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AERODYNAMIC CHARACTERISTICS  
OF A PROPULSIVE WING/CANARD  
CONCEPT AT STOL SPEEDS

V. R. Stewart

Rockwell International Corporation  
Columbus, Ohio 43216

Contract NAS1-17171

November 1985

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National Aeronautics and  
Space Administration

Langley Research Center  
Hampton, Virginia 23665



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## ABSTRACT

A full span model of a wing/canard concept representing a fighter configuration has been tested at STOL conditions in the NASA Langley 4 by 7 Meter Tunnel. The results of this test are presented, and comparisons are made to previous data of the same configuration tested as a semispan model. The potential of the propulsive wing/canard to develop very high lift coefficients was investigated with several nozzle spans (nozzle aspect ratios). Although longitudinal trim was not accomplished with the blowing distributions and configurations tested, the propulsive wing/canard appears to offer an approach to managing the large negative pitching moments associated with trailing edge flap blowing. Also presented are data showing the effects of large flap deflections and relative wing/canard positions. Presented in the appendix to the report are limited lateral-directional and ground effects data, as well as wing downwash measurements.

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## SYMBOLS

	A	~	Aspect Ratio	~ $b^2/S$
	AF	~	Aft Force - balance	
	b	~	Span	
	b <sub>exposed</sub>	~	Exposed Span	
	b <sub>f</sub>	~	Flap Span	
BN/B	b <sub>j</sub>	~	Jet Span/Exposed Span	
	BP	~	Butt Plane (B.P. = 0 at wing root)	
	$\bar{c}$	~	Mean Aerodynamic Chord	
CD	C <sub>D</sub>	~	Drag Coefficient	~ $\frac{\text{Drag}}{qS}$
CDTR	C <sub>D<sub>TR</sub></sub>	~	Thrust Removed Drag Coefficient	
CL	C <sub>L</sub>	~	Lift Coefficient	~ $\frac{\text{Lift}}{qS}$
CLTR	C <sub>L<sub>TR</sub></sub>	~	Thrust Removed Lift Coefficient	
CM	C <sub>M</sub>	~	Pitching Moment Coefficient	~ $\frac{\text{Pitching Moment}}{qS \bar{c}}$
CMTR	C <sub>M<sub>TR</sub></sub>	~	Thrust Removed Pitching Moment Coefficient	
	C <sub>p</sub>	~	Pressure Coefficient	
CMU	C <sub>μ</sub>	~	Blowing Coefficient	~ $\frac{\dot{m} V_j}{qS}$
CN	C <sub>n</sub>	~	Yawing Moment Coefficient	
CROLL	C <sub>ℓ</sub>	~	Rolling Moment Coefficient	
CY	C <sub>y</sub>	~	Side Force Coefficient	
	FRL	~	Fuselage Reference Line	
	H	~	Height of FRL above Ground	
	K <sub>b</sub>	~	Span Correlation Factor	



## SYMBOLS (Concluded)

	<b>M</b>	~	Mach Number
	$\dot{m}$	~	Nozzle Mass Flow
	<b>NF</b>	~	Normal Force - Balance
	$P_L$	~	Local Static Pressure
	$P_\infty$	~	Ambient Pressure
	<b>q</b>	~	Dynamic Pressure ~ $\frac{1}{2}\rho V^2$
	<b>S</b>	~	Reference Area
$V_\infty$	<b>V</b>	~	Freestream Velocity
	$V_j$	~	Jet Velocity
	<b>Y</b>	~	Spanwise Measurement
	$\alpha$	~	Angle of Attack of Fuselage
	$\rho$	~	Density of Air
DELC	$\delta_C$	~	Canard Deflection
DELF	$\delta_F$	~	Flap Deflection
	$\epsilon$	~	Downwash Angle

### Subscripts

<b>c</b>	~	Canard
<b>N</b>	~	Nozzle
<b>w</b>	~	Wing
<b>TR</b>	~	Thrust Removed

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## 1.0 INTRODUCTION

Attainment of very short ground roll distances (approximately 400 feet) for takeoff and landing requires that operation at flight speeds of approximately 50 knots must be attainable. In today's fighter design, high thrust to weight ratios are required for maneuvering and high speed flight. If a concept can be developed which can utilize these thrust levels to augment the aerodynamic lift, then it may be possible to operate at these speeds.

One such concept involves the use of all or a large percentage of the engine exhaust ducted over the flap of the lifting surfaces. The blown flap or propulsive wing has been recognized as a method of producing large circulation lift coefficients. However, these large induced circulation lift coefficients, as well as the deflected thrust vector, react well aft on the wing and produce sizable nose-down pitching moments. The addition of a propulsive canard offers a possible conceptual design relief for these sizable trim requirements. The displacement of a portion of the thrust forward, as well as the induced circulation lift on the canard, tend to balance the moment due to the aft loading on the wing.

An earlier program done on this propulsive wing/canard configuration was primarily a cruise and maneuvering investigation. During that study (ref. 1), it was demonstrated that the concept would provide an improved maneuvering lift/drag (L/D) at transonic speeds. Also, the earlier study indicated quite large circulation lift coefficients at relatively low flap deflections. The study was done with a semispan model and tests were conducted in the Ames 14 foot Transonic Tunnel and in the Rockwell 7 by 10 foot low speed tunnel. Flap deflections in these tests were limited to a maximum of 15 degrees. The major configuration variable investigated was the effect of wing/canard relative locations. Analysis of that data showed that the best position for the low flap deflection ( $\delta_F \leq 15$  degrees) and for all speeds ( $M = 0.1$  to  $0.9$ ) was with the canard high relative to the wing. These test results were discussed in References 1 through 3.

The investigation of the propulsive wing/canard concept has continued with additional tests and analysis. The data have been extended to the higher flap deflections and into ground effects as required for STOL operation, see Reference 4.

This report covers this test phase of the propulsive wing/canard concept investigation in STOL conditions in the NASA Langley 4 by 7 Meter Tunnel. The major variables investigated were:

- (1) blown flap span
- (2) canard/wing relative location, and
- (3) flap deflection with large blowing coefficients.

## 2.0 MODEL AND TEST PROCEDURE

### 2.1 Model Description

The model is a full span wing/canard configuration. Both wing and canard have blown trailing edge flaps which can be deflected from zero to 60 degrees. The nozzle slots are at the flap hinge line (the 80 percent local chord position) and are perpendicular to the fuselage centerline. Figures 1 and 2 are model sketches presenting dimensional data. Table 1 presents a tabulation of the model geometry. The canard can be placed in one of three positions on the fuselage, and the wing can be placed in one of two positions on the fuselage, as shown in Figure 1. Figure 1 also shows the location of the downwash probe mounted one mean aerodynamic chord behind the wing.

The span of the nozzle slot on the wing was also variable. Provisions were made for the nozzle to blow either full span, half span, or a quarter span of the flap while maintaining approximately the same nozzle exit area. The nozzle was also always on the inboard portion of the wing. The canard nozzle was similarly configured except that only full span or half span configurations could be tested. When flap deflections were tested, the flap was deflected as a full span flap regardless of the extent of nozzle span. Wing and canard airfoil coordinates are tabulated in Table 2.

Air for the blowing slots is introduced to the model through a pressure reducer valve to the main fuselage plenum (see Figure 3). From the main fuselage plenum, air is ducted to four smaller plenums in the fuselage, one each for two canards and two wings.

Figure 3 also shows the balance installation and the manner in which air is supplied across the balance. The wings and canards plug into the fuselage plenums, allowing the air to flow to the wing or canard high pressure plenums where the flow is stagnated and ducted through pressure drop supply ducts to the low pressure plenums which supply the nozzle slots. The air supply system provided nearly uniform jet exit velocity with span. Figure 4 is a sketch showing the model air supply from the LaRC supply pipe to the nozzle exits. Flow split is adjustable to each of the four blowing surfaces by use of valves located on the main fuselage plenum. High pressure air brought onto the model in this manner results in balance constants which are determined by calibration with the air pipe in place and by a pressure tare which is a function of model internal geometry. These calibrations and tares are a part of the standard, NASA provided, data reduction capability.

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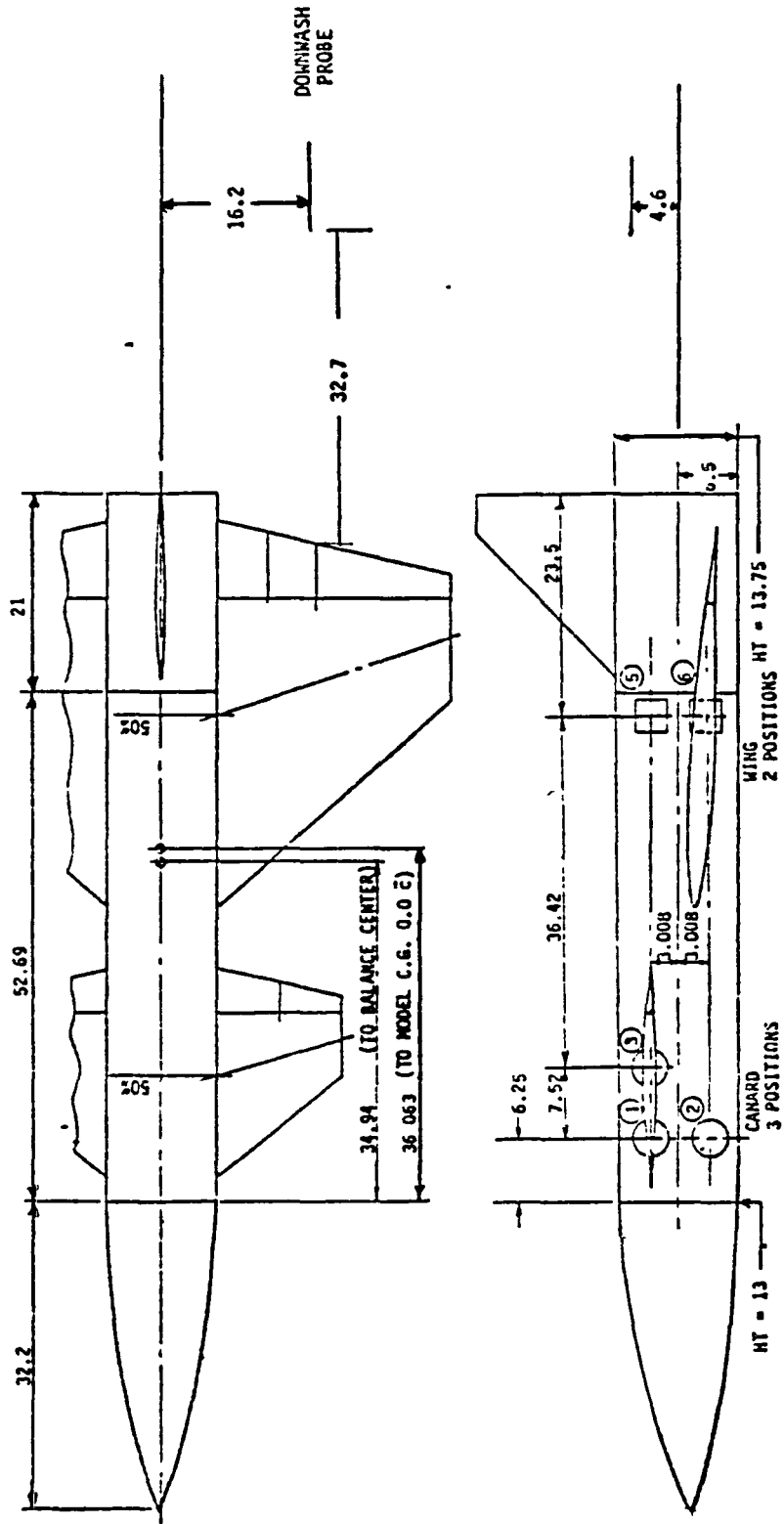


Figure 1. Model Sketch, Surface Locations

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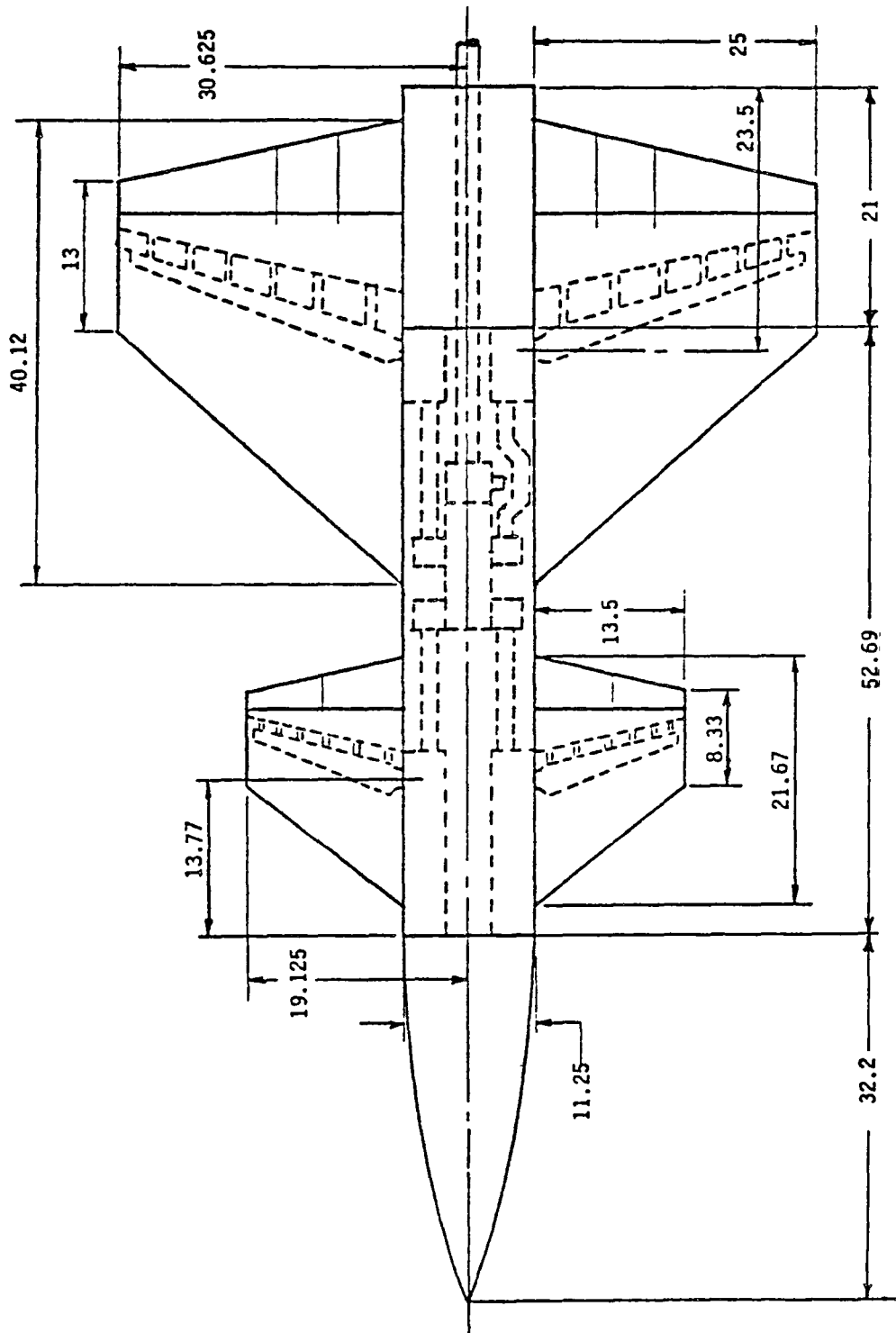


Figure 2. Model Sketch (Dimensions)

TABLE 1. - MODEL GEOMETRY

	WING	CANARD	BODY	TAIL
TIP CHORD	13 IN	8.33 IN	-	4.25 IN
ROOT CHORD EXPOSED	40.12 IN	21.67 IN	-	19.50 IN
ROOT CHORD TOTAL (BP 0)	46.21 IN	27.23 IN	-	-
TAIL HEIGHT	-	-	-	15.20 IN
SPAN TOTAL	61.25 IN	38.25 IN	-	-
AREA EXPOSED	9.222 FT <sup>2</sup>	2.812 FT <sup>2</sup>	-	1.25 FT <sup>2</sup>
AREA TOTAL	12.591 FT <sup>2</sup>	3.334 FT <sup>2</sup>	-	-
ASPECT RATIO EXPOSED	-	1.80	-	2.56
ASPECT RATIO TOTAL	2.069	3.047	-	-
BODY LENGTH	-	-	105.89 IN	-
BODY WIDTH	-	-	11.25 IN	-
MINIMUM BODY HEIGHT	-	-	13.00 IN	-
MAXIMUM BODY HEIGHT	-	-	13.75 IN	-
MAC EXPOSED	28.87 IN	16.0 IN	-	-
MAC TOTAL	32.71 IN	19.45 IN	-	-
SWEEP	41 DEG	38.33 DEG	-	45 DEG

TABLE 2. - BASIC AIRFOIL COORDINATES (NO NOZZLE)

X/C	WING				CANARD			
	NO DROOP		DROOP		NO DROOP		DROOP	
	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER
0	0	0	-.035	-.035	0	0	-.025	-.025
.002	.0046	-.0044	-.0299	-.0395	.0046	-.0044	-.0195	-.0295
.005	.0068	-.00605	-.0263	-.04095	.0068	-.00605	-.0154	-.0311
.01	.0093	-.0077	-.0217	-.0411	.0093	-.0077	-.01035	-.0308
.02	.0126	-.0100	-.0150	-.0398	.0126	-.0100	-.003	-.029
.03	.0153	-.0120	-.0097	-.0383	.0153	-.0120	.0024	-.0278
.04	.0175	-.0133	-.0051	-.0370	.0175	-.0133	.007	-.0266
.06	.0212	-.0157	.0034	-.0346	.0212	-.0157	.0142	-.0249
.08	.0242	-.0176	.0107	-.0324	.0242	-.0176	.01955	-.0235
.10	.0264	-.0192	.0165	-.0303	.0264	-.0192	.0235	-.0226
.125	.0287	-.0208	.02235	-.0279	.0287	-.0208	.0273	-.0220
.15	.0305	-.0222	.0267	-.0258	.0305	-.0222	.030	-.0224
.20	.0329	-.0241	.0321	-.0246	.0329	-.0241	.0329	-.0241
.25	.0342	-.0254	.0342	-.0254	.0342	-.0254	.0342	-.0254
.30	.0350	-.0256	.0350	-.0256	.0350	-.0256	.0350	-.0256
.35	.03548	-.02545	.03548	-.02545	.0354	-.02545	.0354	-.02545
.40	.0357	-.0249	.0357	-.0249	.0357	-.0249	.0357	-.0249
.45	.03575	-.0241	.03575	-.0241	.0358	-.0241	.0358	-.0241
.50	.03565	-.0230	.03565	-.0230	.0358	-.0230	.0358	-.0230

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TABLE 2. - BASIC AIRFOIL COORDINATES (NO NOZZLE) (Concluded)

X/C	WING				CANARD			
	NO DROOP		DROOP		NO DROOP		DROOP	
	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER
.55	.03535	-.02175	.03535	-.02175	.0358	-.02175	.0358	-.02175
.60	.03488	-.01945	.03488	-.01945	.0356	-.01945	.0356	-.01945
.65	.0342	-.0165	.0342	-.0165	.03535	-.0165	.03535	-.0165
.70	.0332	-.0126	.0332	-.0126	.0348	-.0126	.0348	-.0126
.75	.03165	-.0081	.03165	-.0081	.0340	-.0081	.0340	-.0081
.80	.029	-.0028	.0290	-.0028	.0325	-.0028	.0325	-.0028
.85	.02325	+.002	.02325	+.002	.02325	+.002	.02325	+.002
.90	.0173	+.003	.0173	+.003	.0176	+.003	.0176	+.003
.95	.00935	+.0008	.00935	+.0008	.0112	+.0008	.0112	+.0008
1.00	0	-.004	0	-.004	.004	-.004	.004	-.004

FLAP

L.E. RADIUS = .012

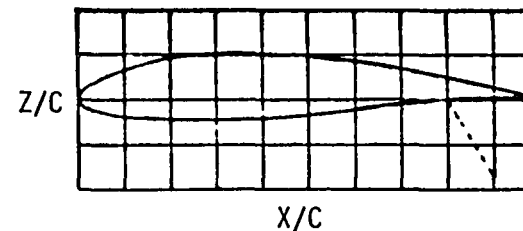
WING ROOT INCIDENCE +4.0°

WING TIP INCIDENCE -1.0°

TWIST ALONG X/C = 0.80

Z ~ VERTICAL DISTANCE

C ~ CHORD ~ IN.



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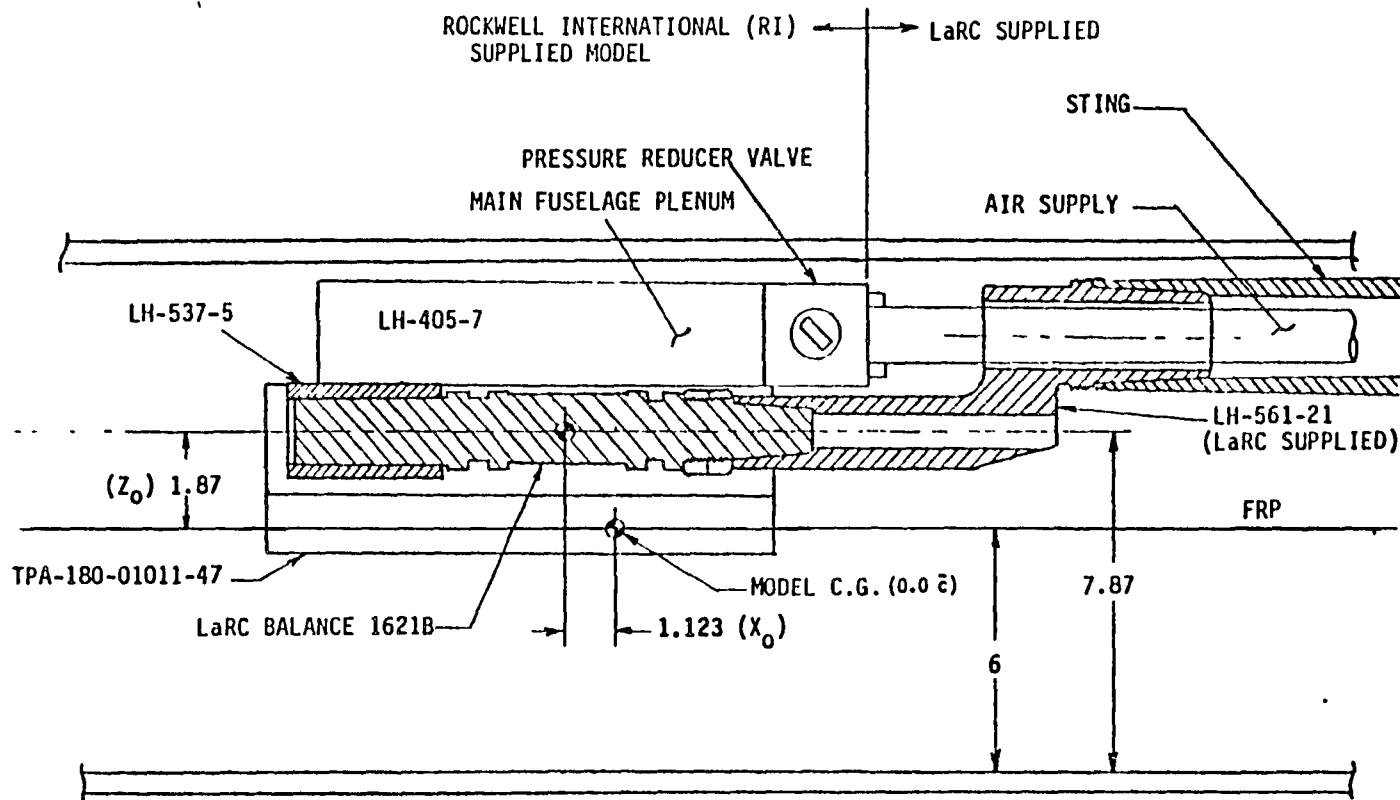


Figure 3. Schematic of Sting, Balance, and Air Supply

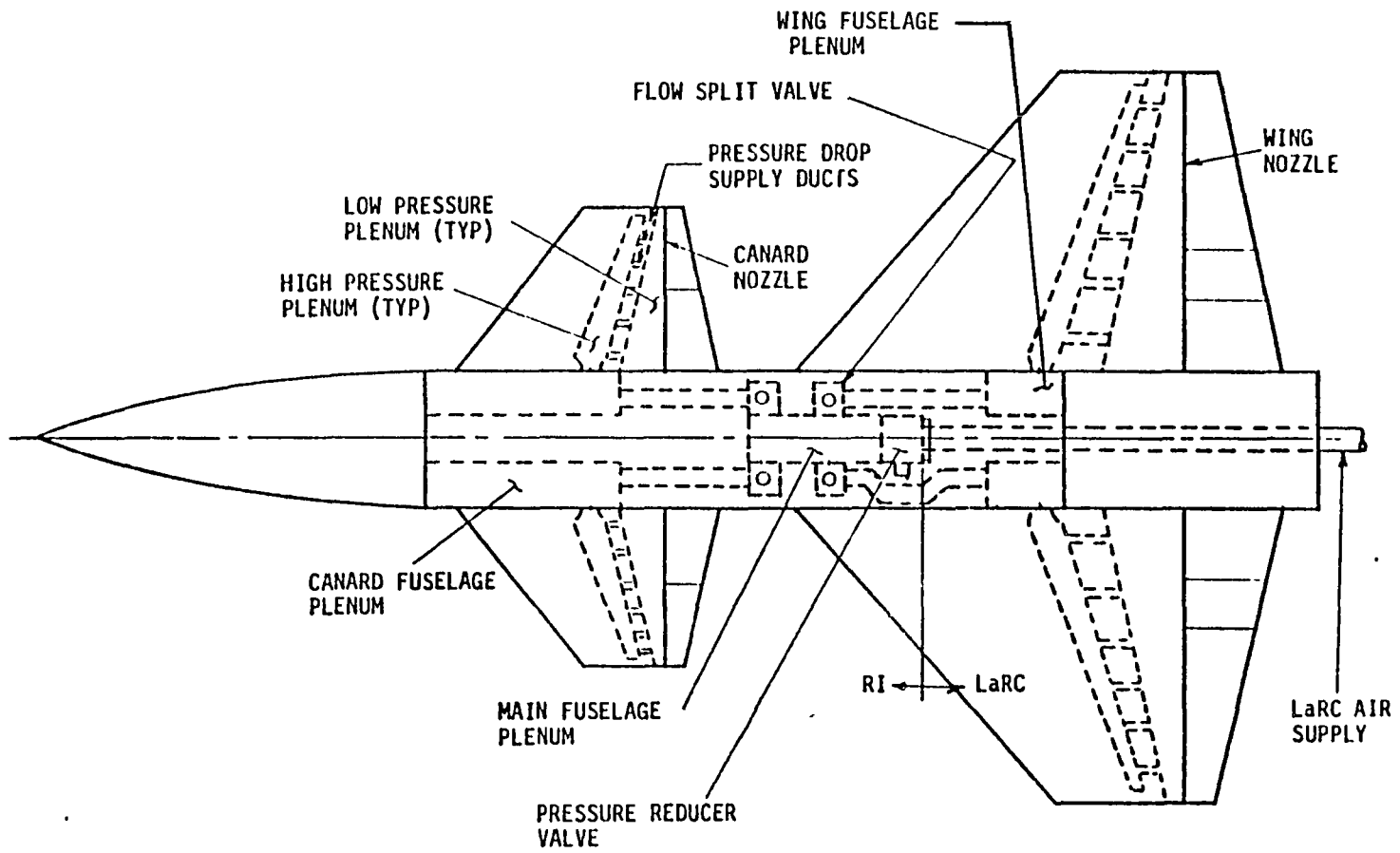


Figure 4. Model Air Supply Diagram

## 2.2 Model Instrumentation

The model and test instrumentation consisted of the following:

1. six-component internal balance
2. wing surface static pressure ports
3. canard surface static pressure ports
4. fuselage surface static pressure ports
5. internal flow pressure sensors
6. airflow measuring and calibrating instrumentation (NASA supplied)
7. calibrated downwash probe (NASA supplied)

Surface pressure instrumentation consisted of a total of 229 surface taps with 151 located on the left hand wing, 56 located on the left hand canard, and 22 located at 3 inches left of the plane of symmetry on the fuselage upper and lower surfaces. These were all monitored with a standard scani-valve system. Figure 5 shows the location of the wing and canard pressure taps by butt plane and local chord. The locations of the fuselage pressures along the fuselage length are presented with the tabulated pressure data in Appendix A.

The downwash probe was a five-hole directional probe mounted approximately one chord length aft of wing trailing edge (see Figure 1). The calibration of the probe provided local angle of attack relative to model FRL. These angles of attack have been converted to downwash angle and are presented in Appendix A.

## 2.3 Model - Data Reduction

The force data have been reduced to standard six component force and moment coefficient about the stability axis by standard data reduction equations. The blowing coefficient  $C_{\mu}$  is obtained by expanding the measured mass flow to the fully expanded, isentropic, velocity and normalizing on freestream dynamic pressure and the total wing area.

$$C_{\mu} = \frac{\dot{m} V_j}{qS}$$

The thrust removed coefficients are obtained by adjusting the balance raw data output by the static thrust tares and then applying normal corrections to the adjusted data, i.e.:

$$NF_{TR} = NF_{\text{uncorrected}} - NF_{\text{static thrust}}$$

$$AF_{TR} = AF_{\text{uncorrected}} - AF_{\text{static thrust}}$$

etc.

This procedure required that a thrust tare be taken each time a configuration change involving the thrust nozzle system was made.

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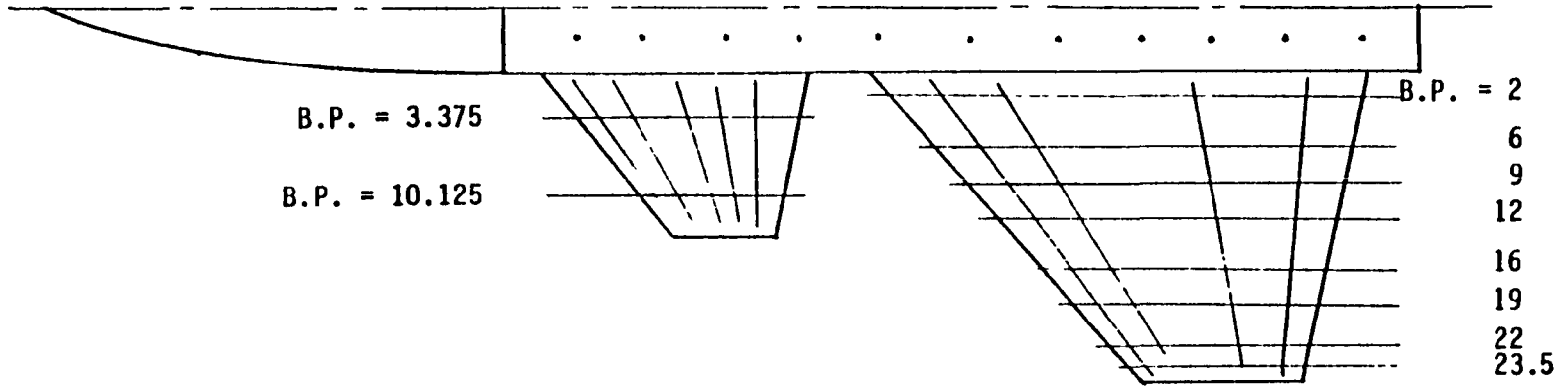


Figure 5. Surface Pressure Tap Locations

CANARD STATIC PRESS. TAP LOCATION				
% Chord	Yc (Dist. to Root Chord)			
	3.375		10.125	
	Up	Lwr	Up	Lwr
0	L.E.		L.F.	
2.5	X	X	X	X
5	X	X	X	X
10	X	X	X	X
15	X	X	X	X
25	X	X	X	X
35	X	X	X	X
50	X	X	X	X
56	X	X	X	X
65	X	X	X	X
73	X	X	X	X
78	X	X	X	X
79	X	X	X	X
80.5	X	X	X	X
81	X	X	X	X
82	X	X	X	X
84	X	X	X	X
87	X	X	X	X
89	X	X	X	X
93	X	X	X	X
96	X	X	X	X
100	T.E.		T.E.	

WING STATIC PRESSURE TAP LOCATIONS																
% Chord	Yw (Distance to Root Chord)															
	2		6		9		12		16		19		22		23.5	
	Up	Lwr	Up	Lwr	Up	Lwr	Up	Lwr	Up	Lwr	Up	Lwr	Up	Lwr	Up	Lwr
0	L.E.		L.E.				L.E.		L.E.				L.E.			
2.5	X	X	X	X			X	X	X	X			X	X		
5	X	X	X	X			X	X	X	X			X	X		
10	X	X	X	X			X	X	X	X			X	X		
15	X	X	X	X			X	X	X	X			X	X		
24	X	X	X	X	X	X			X	X			X	X	X	X
33	X	X	X	X	X	X	X	X	X	X	X	X	X	X	X	X
54	X	X	X	X			X	X	X	X			X	X	X	X
65	X	X	X	X			X	X	X	X			X	X	X	X
73.5	X	X	X	X	X	X			X	X			X	X	X	X
78.5	X	X	X	X			X	X	X	X			X	X	X	X
79.5	X	X	X	X			X	X	X	X			X	X	X	X
80.5	X	X	X	X			X	X	X	X			X	X	X	X
81.25	X	X	X	X			X	X	X	X			X	X	X	X
82	X	X	X	X			X	X	X	X			X	X	X	X
84	X	X	X	X	X	X			X	X			X	X	X	X
87	X	X	X	X			X	X	X	X			X	X	X	X
89	X	X	X	X			X	X	X	X			X	X	X	X
93	X	X	X	X			X	X	X	X			X	X	X	X
96	X	X	X	X			X	X	X	X			X	X	X	X
100	T.E.		T.E.				T.E.		T.E.				T.E.			

Surface pressure distributions were reduced to pressure coefficient by the normal data reduction equations.

$$C_p = \frac{\Delta P}{q} = \frac{P_L - P_\infty}{q}$$

Test data are presented in Appendix A. The force data are plotted, and selected pressure data and downwash angle measurements are tabulated. Table 3 defines the configurations for which data are presented.

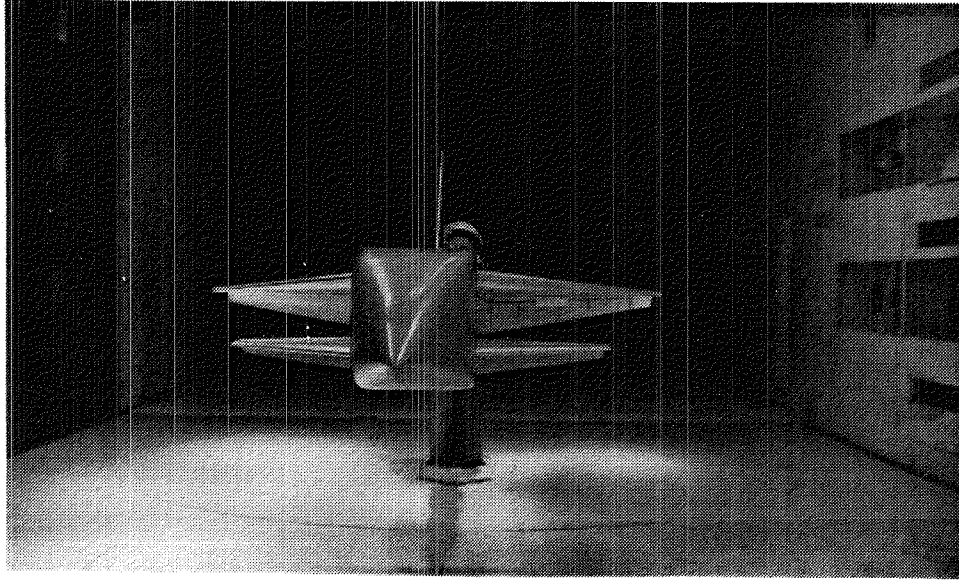
TABLE 3. CONFIGURATION DEFINITION

SYMBOL	DEFINITION	CANARD POSITION	WING POSITION
B OR BV	BODY, VERTICAL	OFF	OFF
BC1V	BODY, CANARD, VERTICAL	1	OFF
BC1W6V	BODY, CANARD, WING, VERTICAL	1	6
BC2W6V	BODY, CANARD, WING, VERTICAL	2	6
BC2W5V	BODY, CANARD, WING, VERTICAL	2	5
BC3W6V	BODY, CANARD, WING, VERTICAL	3	6
BW6V	BODY, WING, VERTICAL	OFF	6

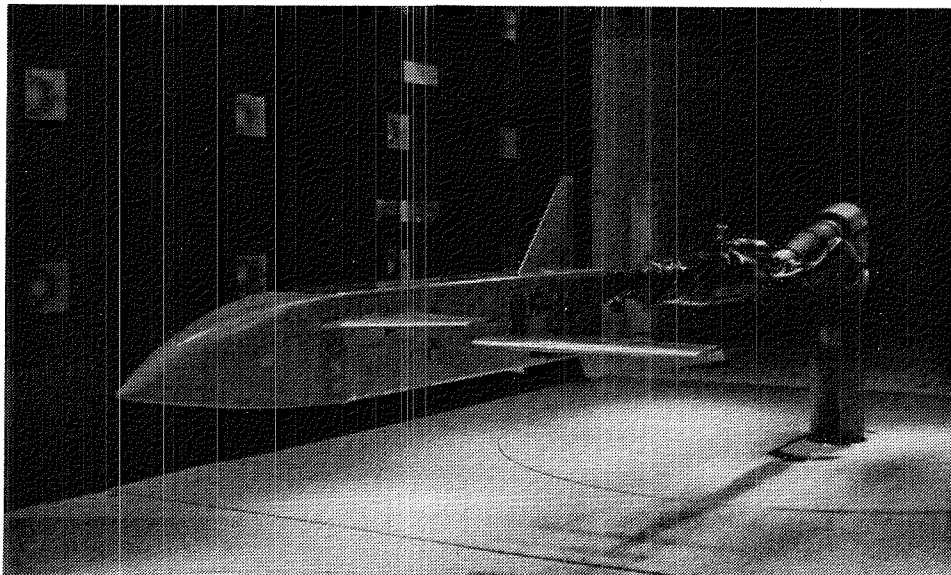
#### 2.4 Installation and Test Procedure

The model is shown installed in the test section of the NASA-Langley 4 x 7 meter tunnel in Figure 6. High pressure air for jet simulation was supplied to the model through an air line in the model support sting, and supplied to each nozzle as described in Section 2.1. The nozzle pressure ratio was maintained at 1.7 or greater, and blowing coefficient was controlled by varying tunnel dynamic pressure. The nozzles were calibrated with the tunnel off, and this static thrust was utilized to obtain the thrust removed coefficients.

The configurations which were tested included: (1) flap deflections of 0, 15, and 45 degrees, (2) blowing spans of 1/4, 1/2, and full span on the wing and 1/2 and full span on the canard, (3) relative wing/canard placement. Blowing coefficients were varied from 0 to 4.0. These configurations were tested both in and out of ground effect. Basic data were obtained through a pitch range of -4 to 20 degrees angle of attack. Selected configurations were tested through a range of yaw angles of +8 to -8 degrees. Where ground effects were investigated, the model was tested at heights from approximately  $h/c = 0.4$  to free air. The sideslip and



(a) Front View



(b) 3/4 Side View

Figure 6. Full Span Model in LaRC 4 x 7 Meter Wind Tunnel Facility

height data were obtained at an angle of attack near zero degrees. Downwash angle was measured behind the wing for most of the configurations. Figure 1 shows the relative wing canard locations of the model, and Figure 5 presents the wing and canard pressure tap locations.

### 3.0 SUMMARY OF TEST RESULTS

A brief analysis of the effect of the blowing nozzle span and wing/canard positioning effects on longitudinal aerodynamic characteristics is presented. The lateral-directional characteristics of the propulsive wing/canard obtained during the test are presented in Appendix A with the complete data. The effect of ground proximity is also presented in Appendix A, A177 to A-209. No unexpected ground effects were noted.

#### 3.1 Effect of Nozzle Span

Nozzle span was varied from approximately 1/4 span to full span with a full span flap deflection of 45 degrees. The effects of nozzle span and blowing coefficient on lift coefficient and thrust removed lift coefficient are presented in Figures 7 through 10. Total  $C_L$  at angles of attack of zero and eight degrees are presented in Figures 7 and 8 with the corresponding thrust removed lift,  $C_{L_{TR}}$ , presented in Figures 9 and 10.

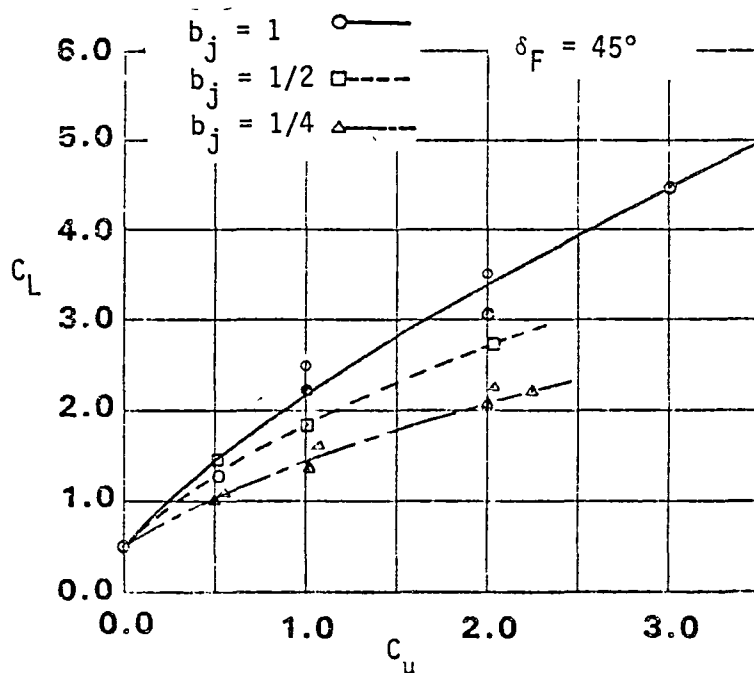


Figure 7. Effect of Blowing Span on Total Lift Coefficient, Canard Off,  $\alpha = 0$  Degrees



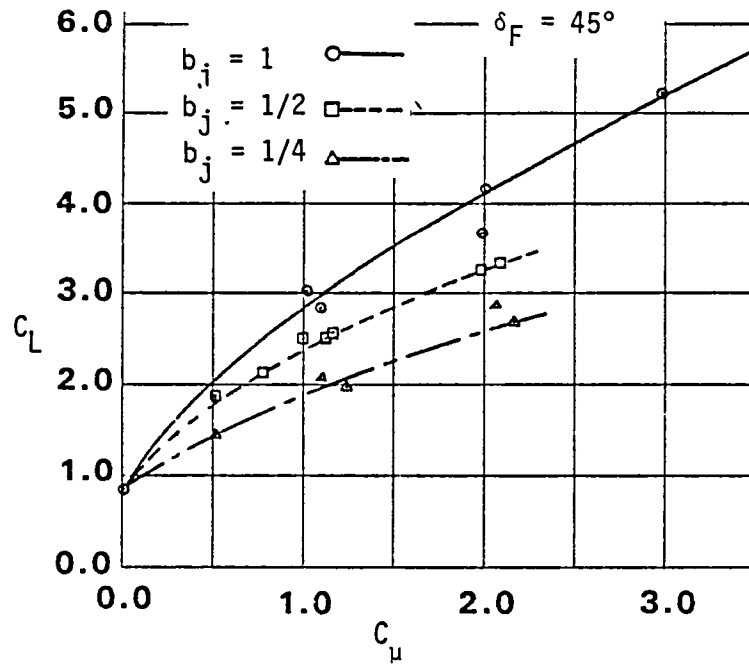


Figure 8. Effect of Blowing Span on Total Lift Coefficient, Canard Off,  $\alpha = 8$  Degrees

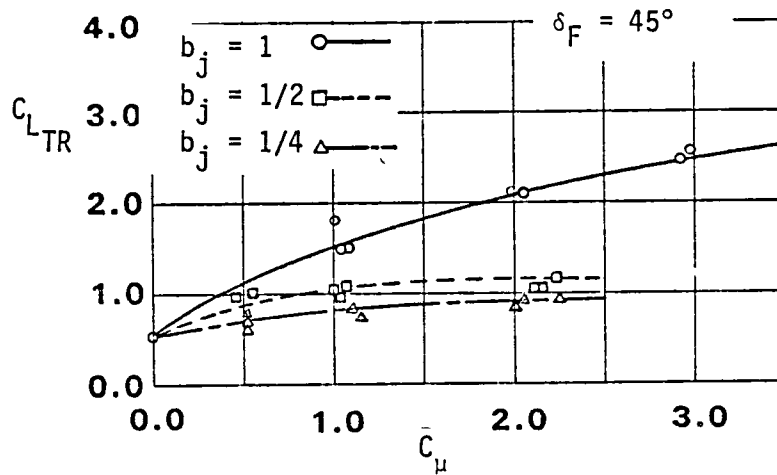


Figure 9. Effect of Blowing Span on the Thrust Removed Lift Coefficient, Canard Off,  $\alpha = 0$  Degrees

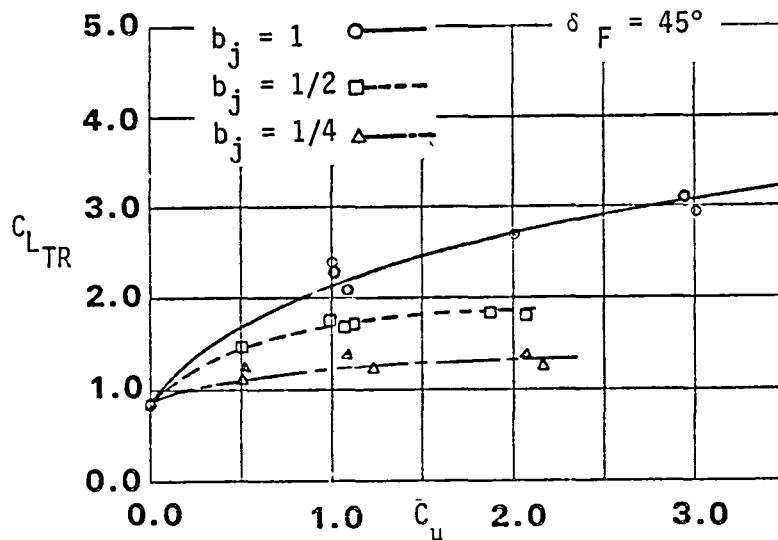


Figure 10. Effect of Blowing Span on the Thrust Removed Lift Coefficient, Canard Off,  $\alpha = 8$  Degrees

These thrust removed lift coefficients are developed by removing the static thrust components from the total force measurements. The results of Figures 9 and 10 show that the full span flap is much superior in developing circulation lift. The full span configuration continues to show an increase in circulation to levels of  $C_{\mu}$  greater than three while the shorter spans appear to have attained their maximum level of circulation lift at a much lower  $C_{\mu}$ . These characteristics are indicative of the greater local blowing coefficient of the shorter nozzles compared to the longer, full span nozzle. The circulation lift increase with blowing is controlled by the local or section blowing coefficient. The jet flap theory accounts for the local  $C_{\mu}$  by the use of the affected wing area ratio, Reference 5.

The drag of the propulsive wing is presented in Figure 11 for the thrust included case and in Figure 12 with the thrust removed. The drag variation demonstrates the same characteristics as the lift. Large induced drag is indicated for the full span nozzle and less induced drag with the shorter span nozzles. This result is to be expected in view of the variation of lift coefficient discussed earlier.

Wing/body pitching moment coefficients presented in Figures 13 and 14 also show the same trend. The moments are about the leading edge of the mean aerodynamic chord; therefore, the vectored thrust inputs a significant negative moment. The thrust removed moment indicates that the aerodynamic load is acting at about the 70 percent chord of the wing. This is not unexpected for a blown flap with a chord of 20 percent of the wing chord.

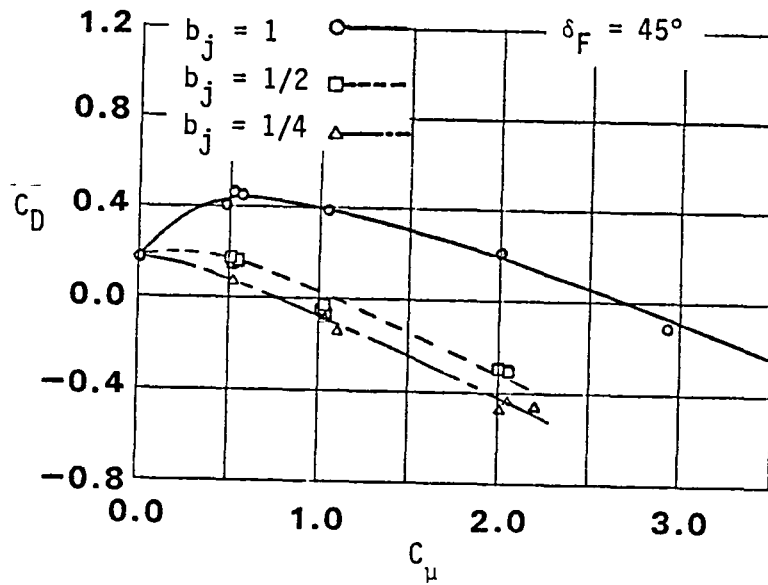


Figure 11. Effect of Blowing Span on the Total Drag Coefficient, Canard Off,  $\alpha = 0$  Degrees

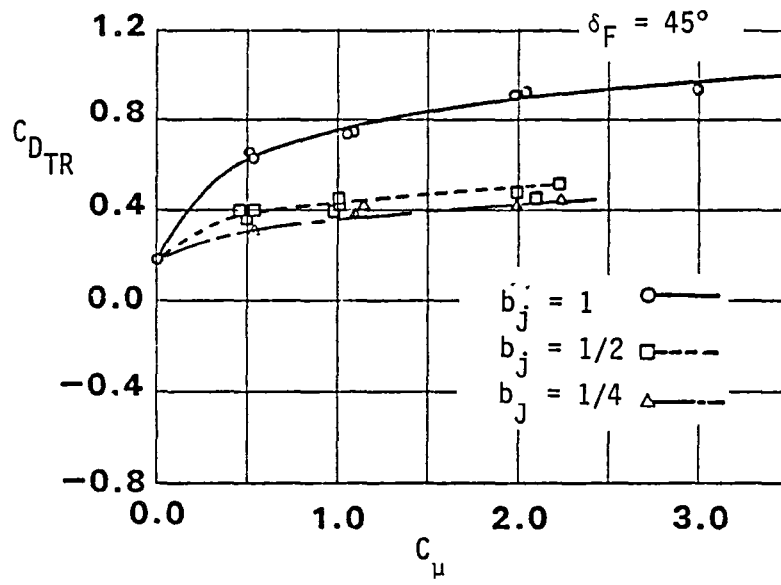


Figure 12. Effect of Blowing Span on the Thrust Removed Drag Coefficient, Canard Off,  $\alpha = 0$  Degrees

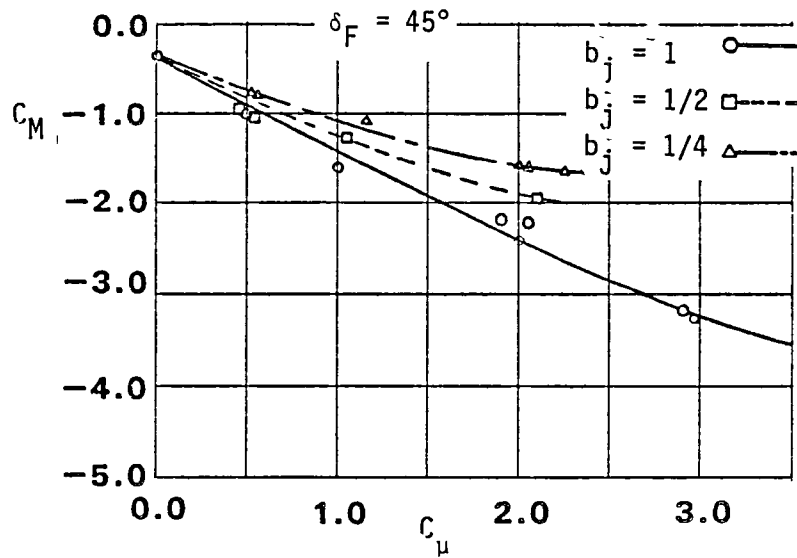


Figure 13. Effect of Blowing Span on the Total Pitching Moment Coefficient, Canard Off,  $\alpha = 0$  Degrees

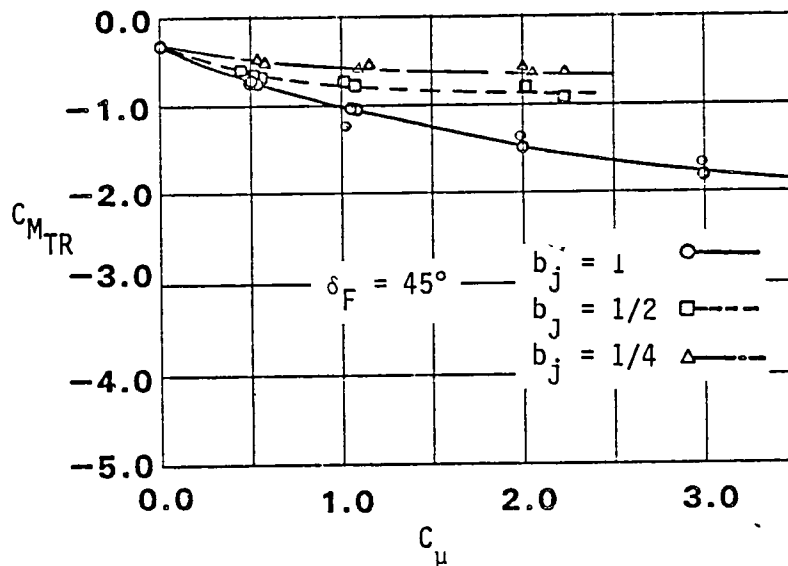


Figure 14. Effect of Blowing Span on the Thrust Removed Pitching Moment Coefficient, Canard Off,  $\alpha = 0$  Degrees

### 3.2 Canard Effects

The negative pitching moments of the propulsive wing require a large force well forward of the center of gravity of the configuration to obtain trim. One means of accomplishing this is using a blown canard. Figure 15 presents the pitching moment coefficient at zero angle of attack with the canard installed in the high forward position and a blowing coefficient of 1/2 that of the wing. Figure 16 presents similar data at an angle of attack of eight degrees. It can be seen that a significant pitch-up moment is gained from the canard, even though it is not sufficient to trim at this center of gravity (CG) location. Increased canard flap deflection, canard incidence angle, or increased canard blowing rates are means of increasing the trim power of the canard. These variables will be addressed in future studies.

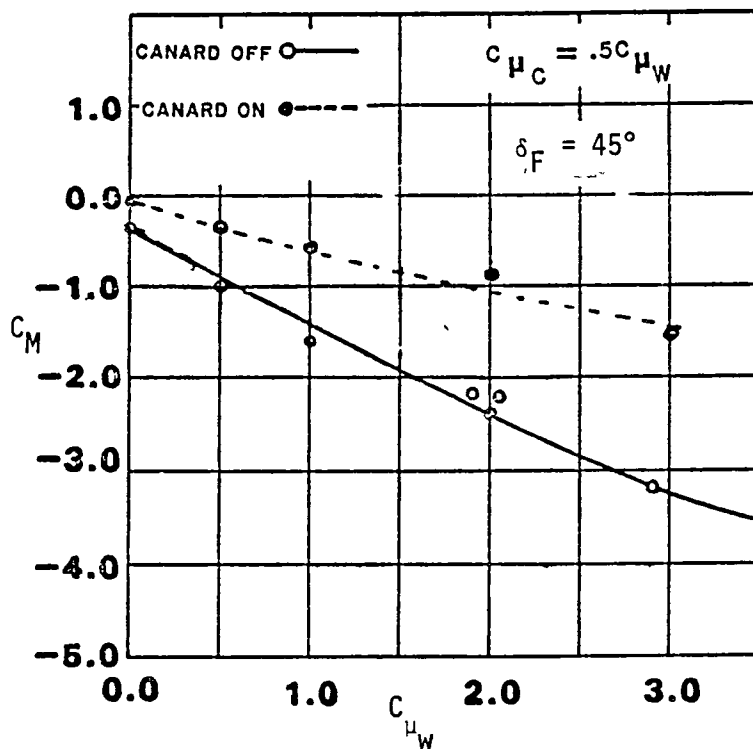


Figure 15. Effect of the Canard on the Pitching Moment Coefficient,  $\alpha = 0$  Degrees

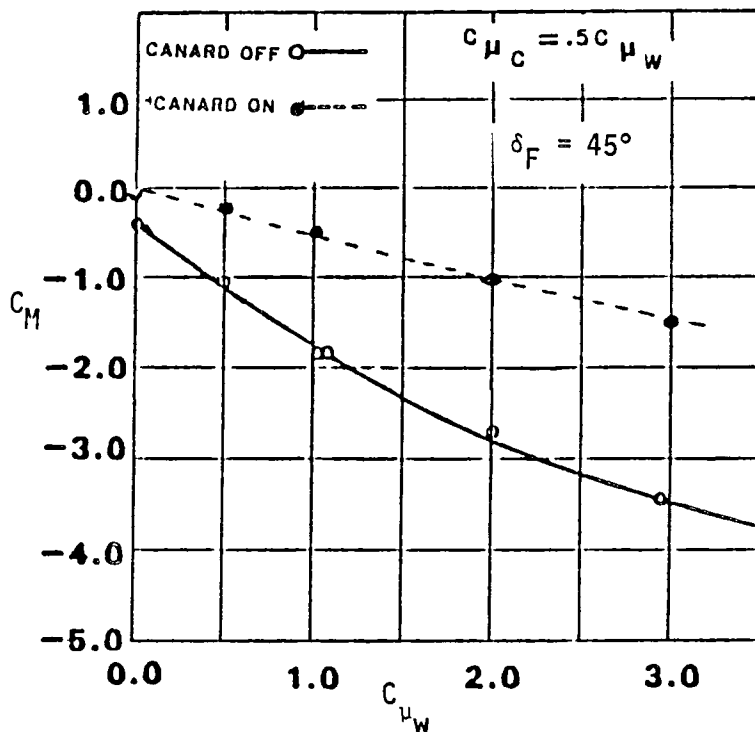


Figure 16. Effect of the Canard on the Pitching Moment Coefficient,  $\alpha = 8$  Degrees

The relative position of the canard and the wing is also a very important consideration in the propulsive wing/canard concept. As is true in most design variables, there is no single position which is the so-called "best" for all conditions. Previous studies (Refs. 1, 2, and 3) showed that for deflections of 15 degrees and less, the high canard, low wing had the better characteristics. This held for blowing as well as for the non-blowing conditions. Early, unpublished, studies on a similar configuration indicated that at nozzle deflections of 60 degrees, the canard should be low relative to the wing. The results of a positioning variation on the current wing/canard model are presented in Figure 17. The flap deflection for this test was 45 degrees; however, with the upper surface contour aft of the nozzle included, the total jet turning angle was 60 degrees. The data show that for this configuration, the low canard, high wing, position 5,2, has the better lift characteristics for the blown, large deflection conditions.

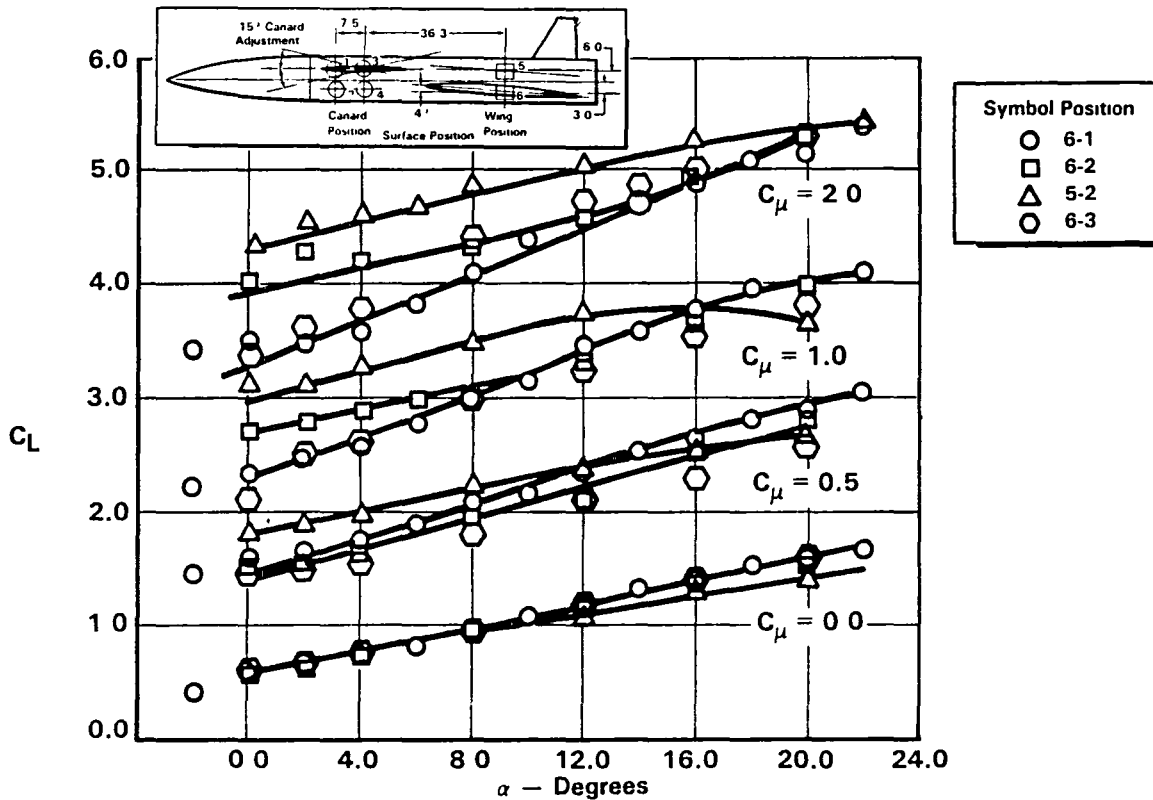


Figure 17. Effect of Canard Position on Total Lift Coefficient

The flight requirements of the STOL wing/canard airplane would dictate the compromises which would finally place the wing and canard in their positions. It is expected that the high speed requirements would prevail, and a propulsive wing/canard configuration would have a high canard with a low wing. Most operations would be at low flap deflections, 15 degrees or less, and the larger deflections would be used only for very short field operations, less than 400 feet.

### 3.3 Semi/Full Span Comparison

The propulsive wing/canard model has been tested in a semispan, as well as a full span configuration. However, the blowing distribution between the wing and canard was changed between the semi and full span configurations, so a comparison of the two configurations powered cannot be made. However, comparisons can be made for the wing alone at blowing coefficients of 0, 1.0, and 2.0 and for the wing/canard configuration with a blowing coefficient of zero.

The comparison of the full and semispan test results for the wing/canard configuration are presented in Figure 18 for the zero blowing case. Very good agreement is seen in most of the parameters. A difference is seen in the lift curve slope, but almost perfect agreement is seen in the drag coefficient. The pitching moment indicates a slightly greater stability for the semispan model. These small differences may have been the result of a change on the canard downwash on the wing.

The comparison of the wing/body is presented in Figure 19 for blowing coefficients of 0, 1.0, and 2.0. Near perfect agreement between the two models is shown in lift and in drag. Again, the semispan model indicates slightly greater stability.

#### 4.0 CONCLUSIONS

The experimental STOL investigation of the propulsive wing/canard has been partially completed. The study, thus far, has shown these results.

The full span nozzle is more effective than the partial span nozzles in producing circulation lift.

The low canard, high wing positioning is preferred for large flap deflection and blowing; however, at other conditions, such as takeoff flaps, cruise, and transonic flight, the high canard is preferred. It is expected that these conditions will prevail, yielding a high canard, low wing configuration.

Very good agreement between full span and semispan model test results were obtained where exact conditions were duplicated.

An additional test of this concept will be conducted with a contoured fuselage. The major objectives with the revised fuselage lines will be to determine the lateral-directional characteristics, to investigate means to trim the wing pitching moments, and to investigate the effect of the moving ground board. Flow field measurements behind the canard will be made for comparison with and as an extension of the semispan model data bank.



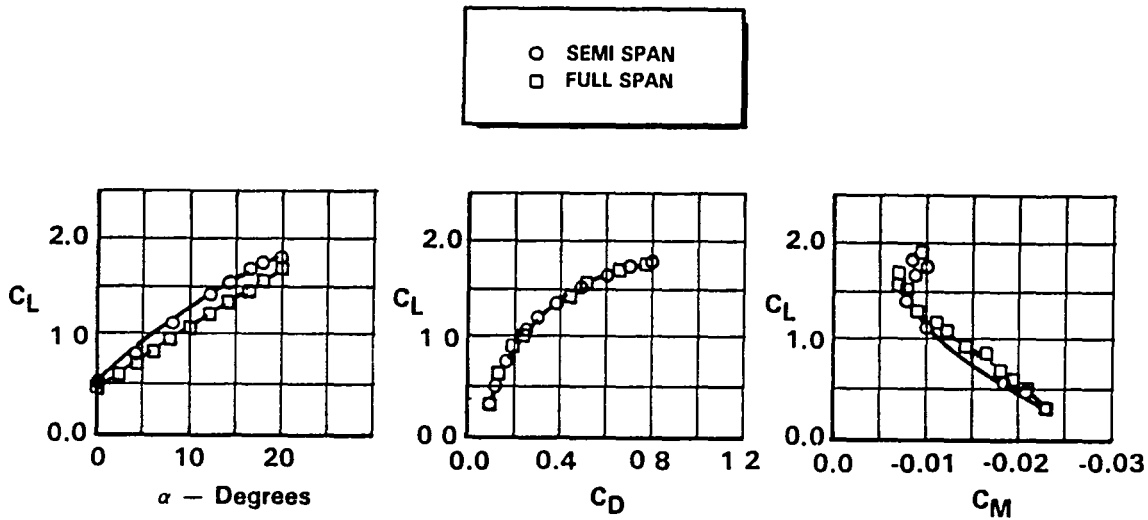


Figure 18. Comparison of Semispan and Full Span Test Results, Wing/Canard/Body,  $C_{\mu} = 0$ ,  $\delta_F = 0$

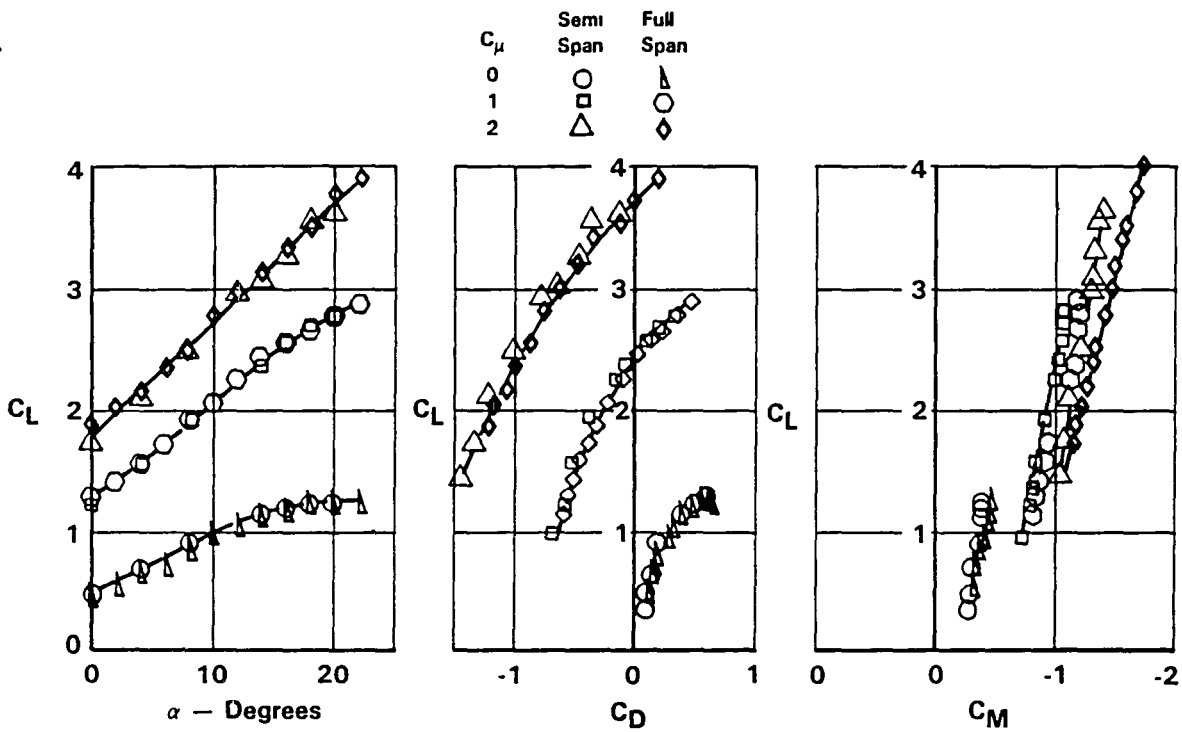


Figure 19. Comparison of Semispan and Full Span Test Results Wing/Body,  $\delta_F = 15$  Degrees

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APPENDIX A

PRESENTATION OF BASIC TEST DATA

APPENDIX A

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## A1.0 INTRODUCTION

The test data are presented in Appendix A. Some mechanical and internal airflow fouling difficulties were experienced during the test period. The air pipe fouling was, apparently, the result of a gradual failure of the air coil internal to the support sting. The air coil eventually ruptured (Run 247), and the sting was replaced with Air Sting #2 ("bent") for the remainder of the test. The mechanical foul appeared to be a combination of several circumstances including: (1) large nose-down moments with the wing alone configuration, (2) an excessively long sting-to-balance adapter coupled with (3) the normal balance flexibility. It is not known if the internal coil failure entered into the mechanical foul, although the initial signs of coil failure did occur prior to the mechanical foul. In any event, those force data runs where a foul can be detected are not presented. In certain cases, surface pressure data and downwash data runs are presented without force data results. These runs represent those for which repeat force runs were made without the pressure instrumentation operating. The surface pressure data and downwash data presented in Appendix A are correct in that the foul experienced did not affect these data. The corresponding force data in some runs may be in error due to the fouling. In these cases, the correct force data run to be used in analysis is noted on the tabulated data.

## A2.0 PLOTTED FORCE DATA

The force and moment data have been plotted and are presented in Section A2.0. The data are presented in three categories. Figures A1 through A24 present the force and moment coefficients for the basic angle of attack variations. Figures A25 through A28 present these coefficients for sideslip variations. And, Figures A29 through A39 show the results of ground height variations. Data are presented for major variables of canard/wing positioning, flap deflection, and flap span. Each major variable is presented as a function of blowing coefficient.

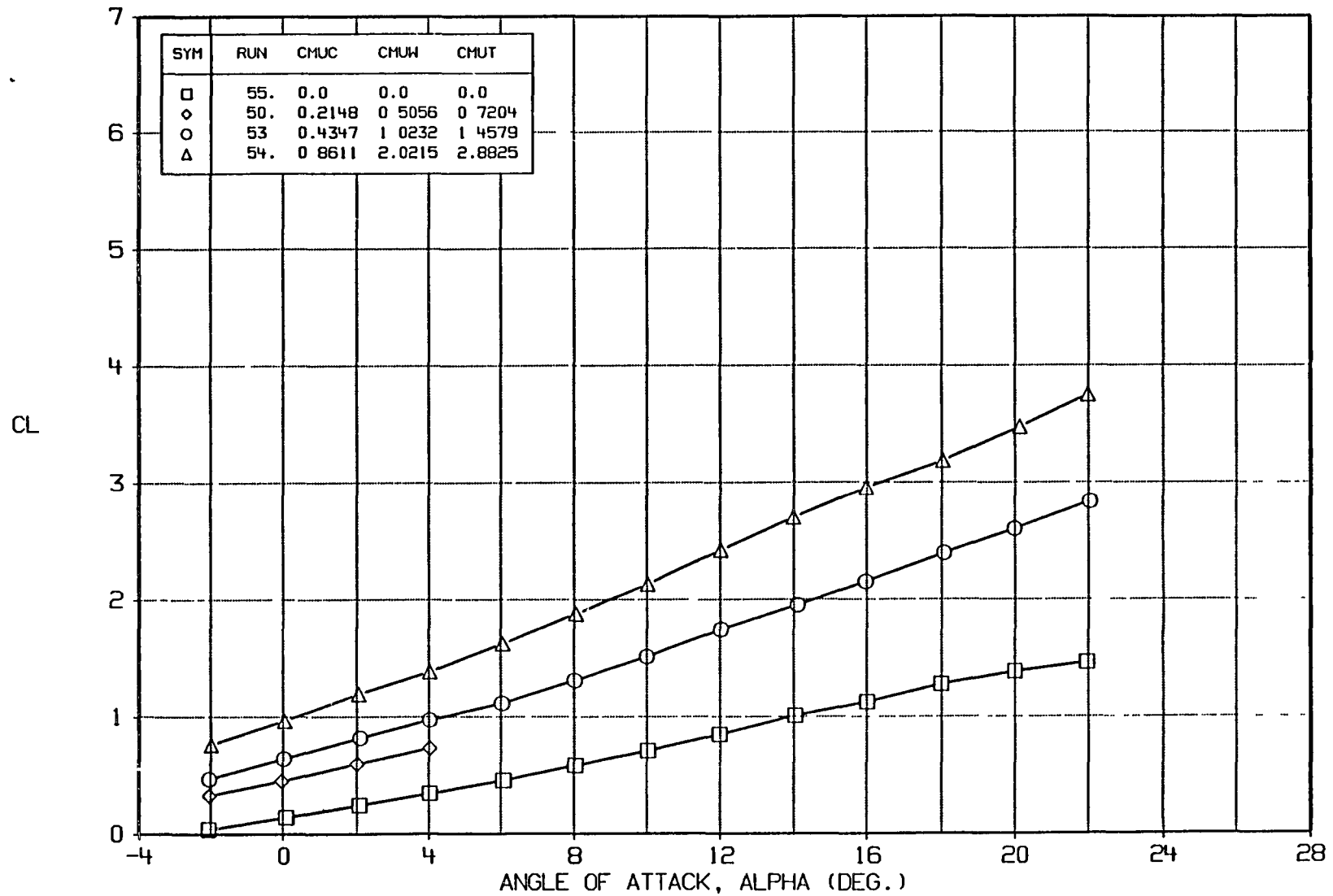


FIGURE A1a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=0, BN/B=1

A-11

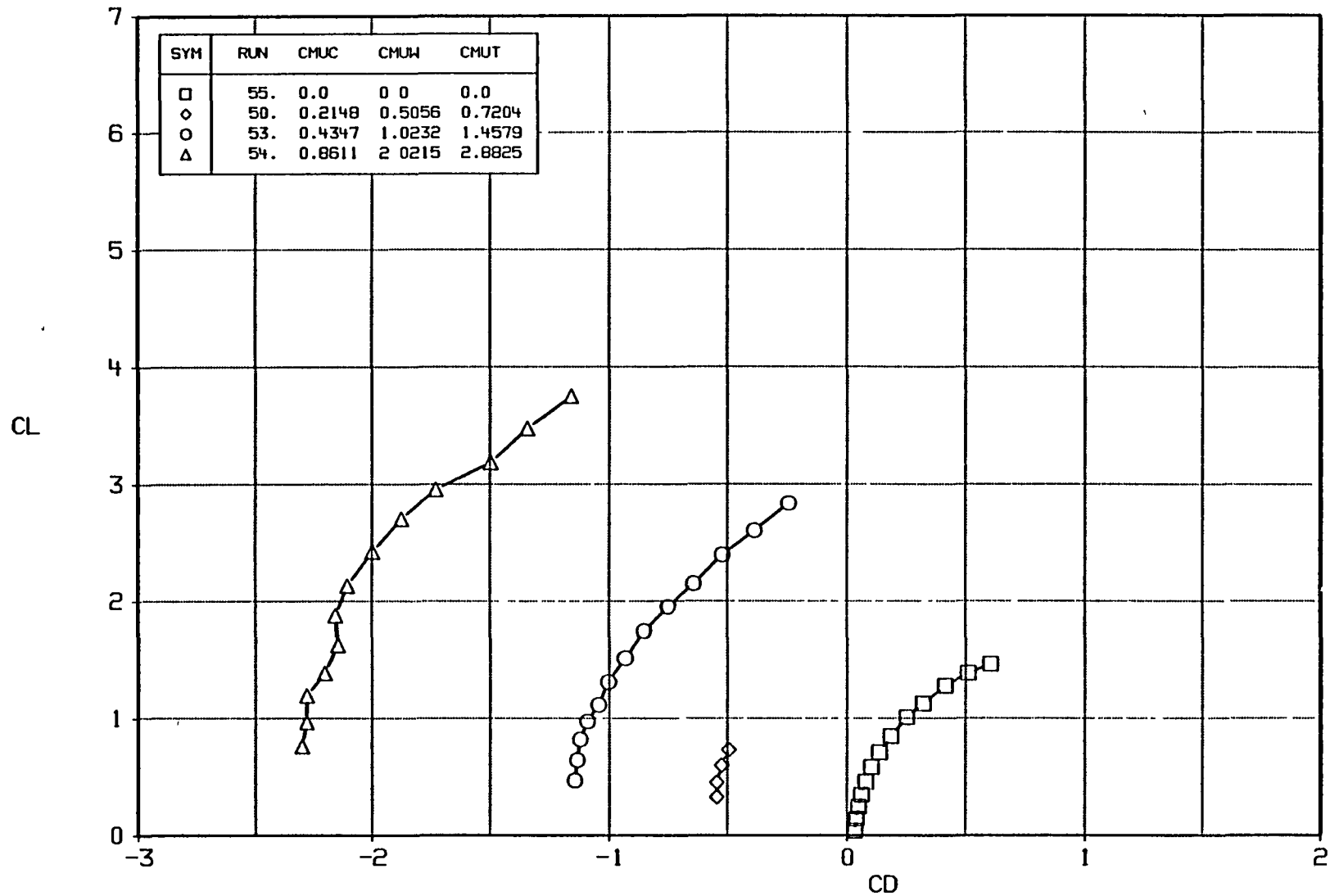


FIGURE A1b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=0, BN/B=1

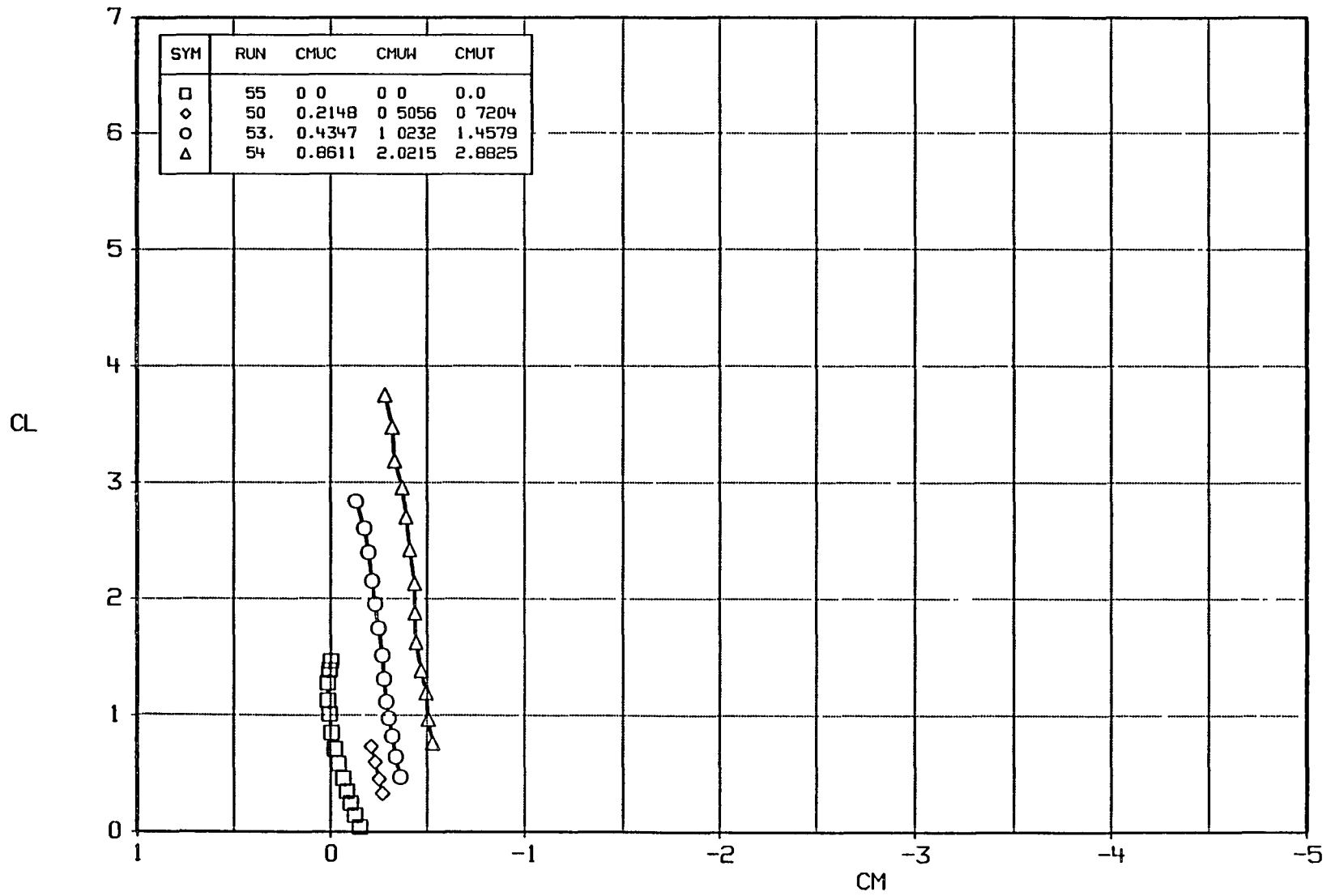


FIGURE A1c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=0, BN/B=1

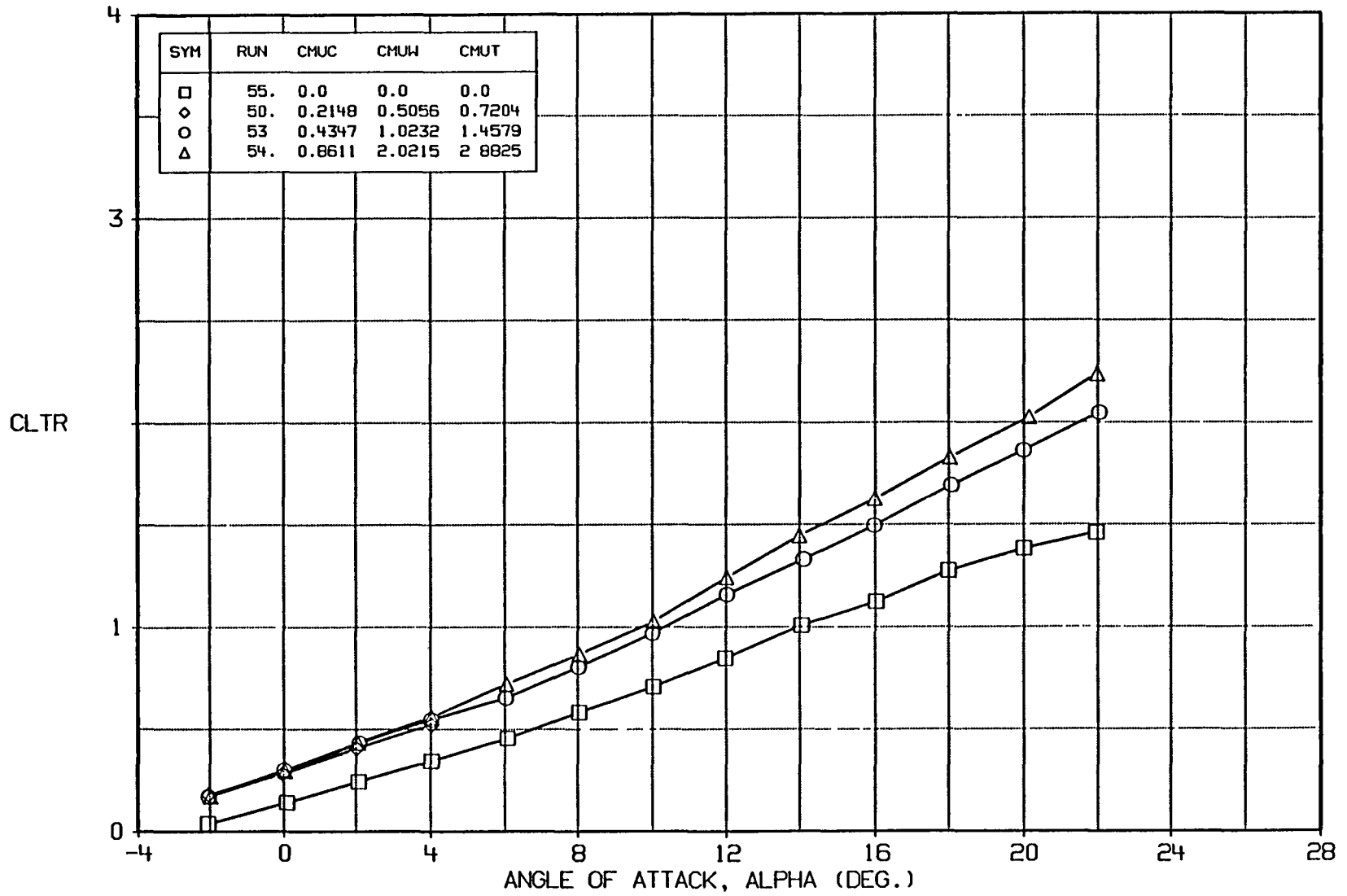


FIGURE A1d BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=0, BN/B=1

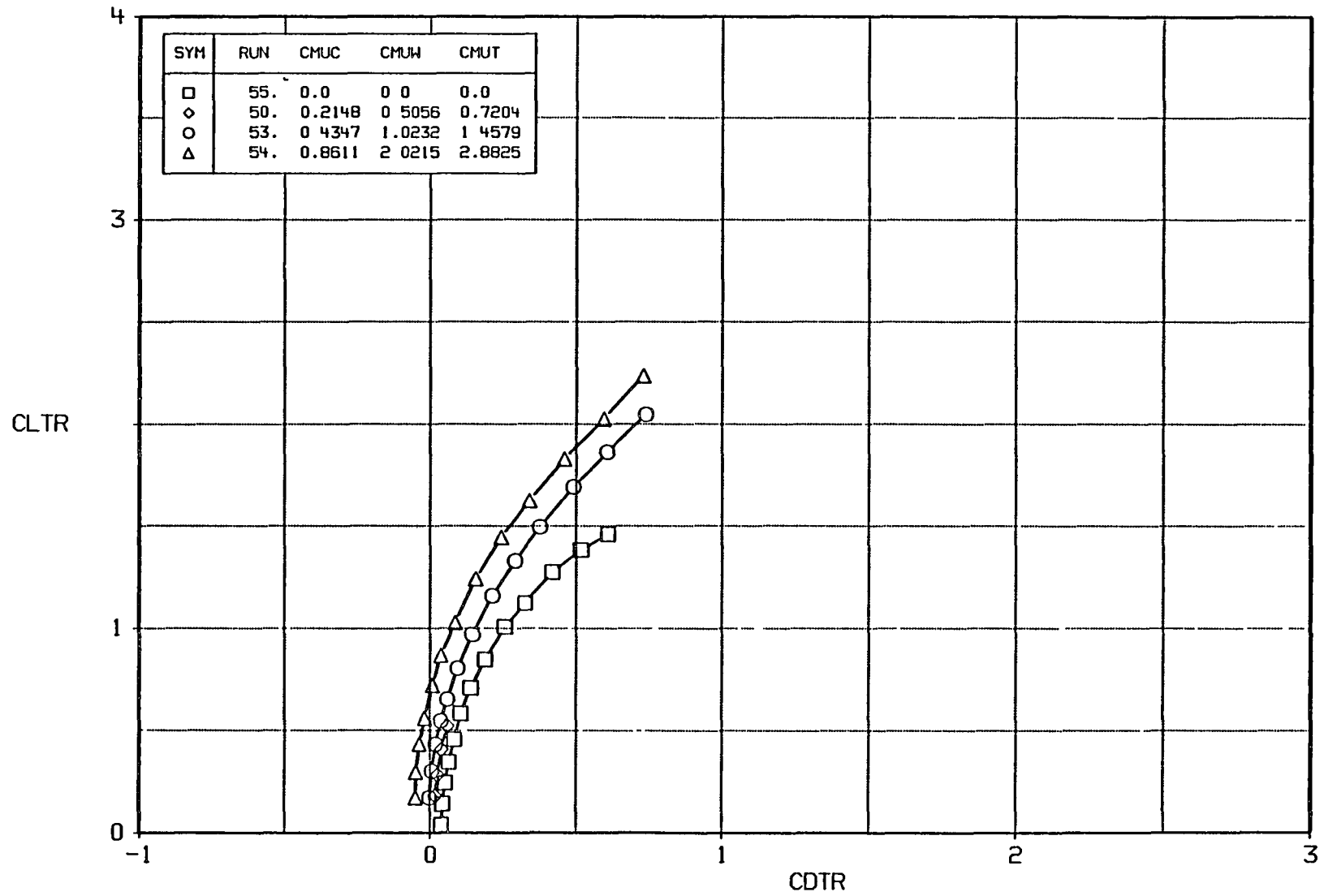


FIGURE A1e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=0, BN/B=1



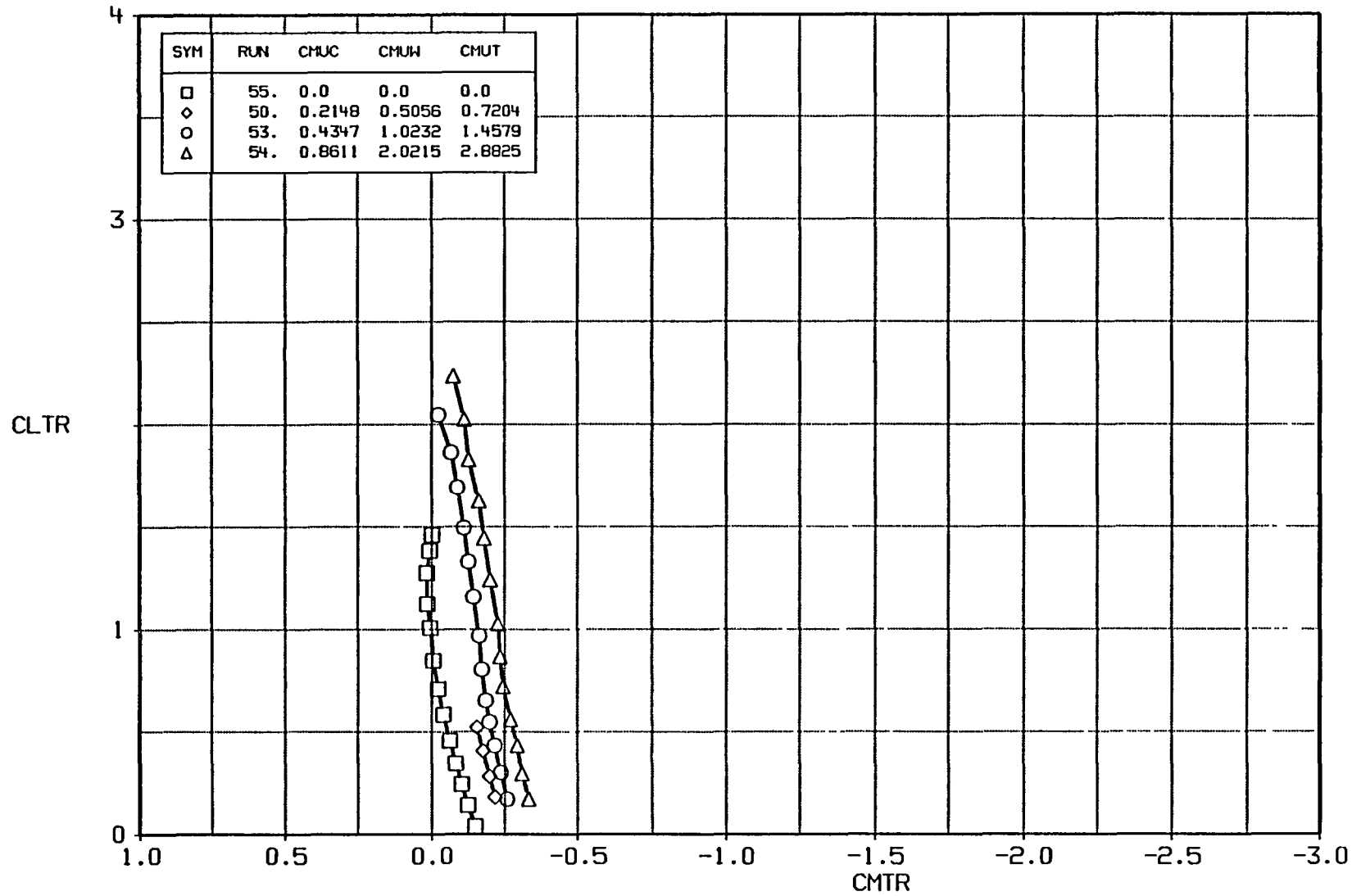


FIGURE A1f BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V, DELF=0, BN/B=1

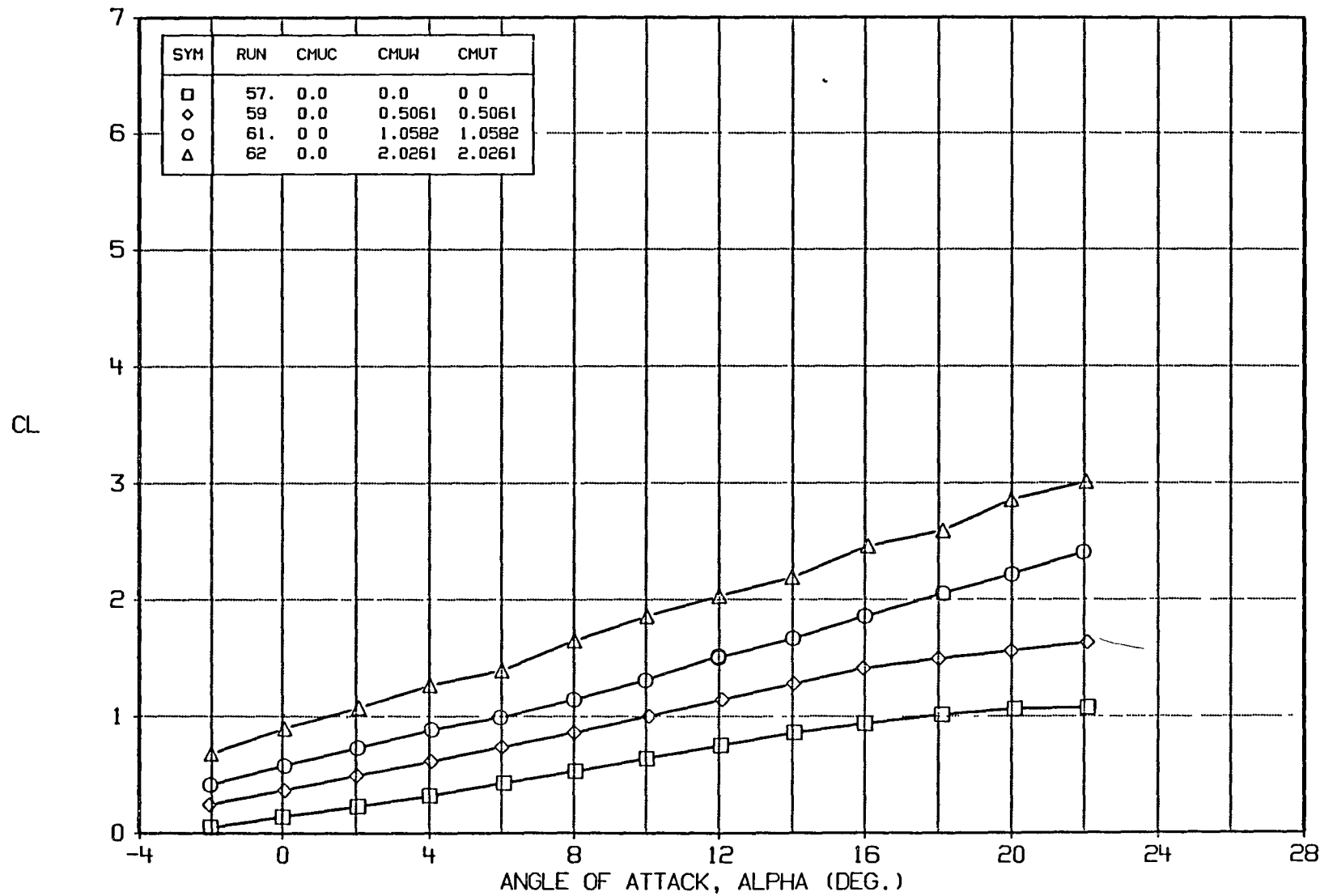


FIGURE A2a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=0, BN/B=1

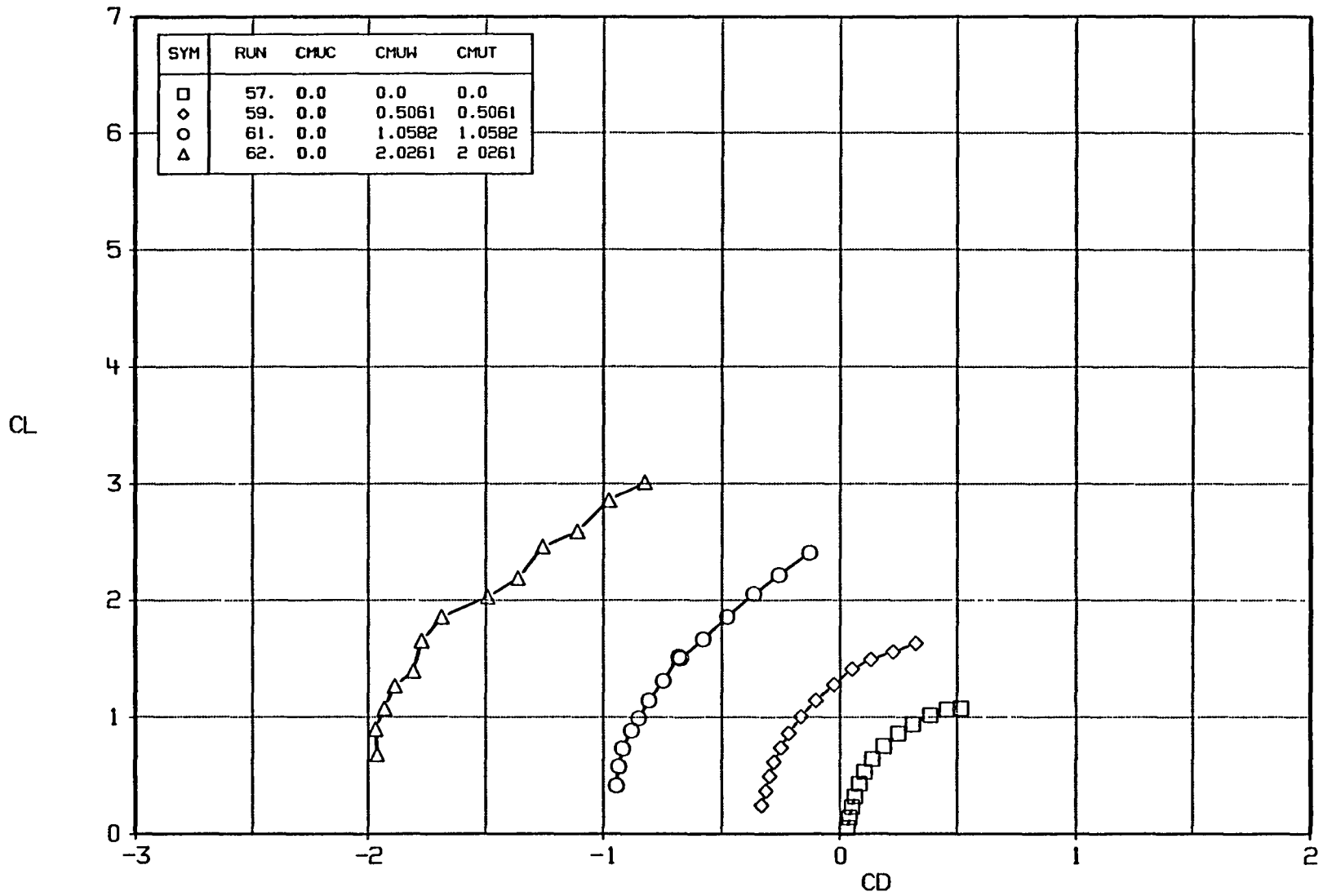


FIGURE A2b BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V, DELF=0, BN/B=1

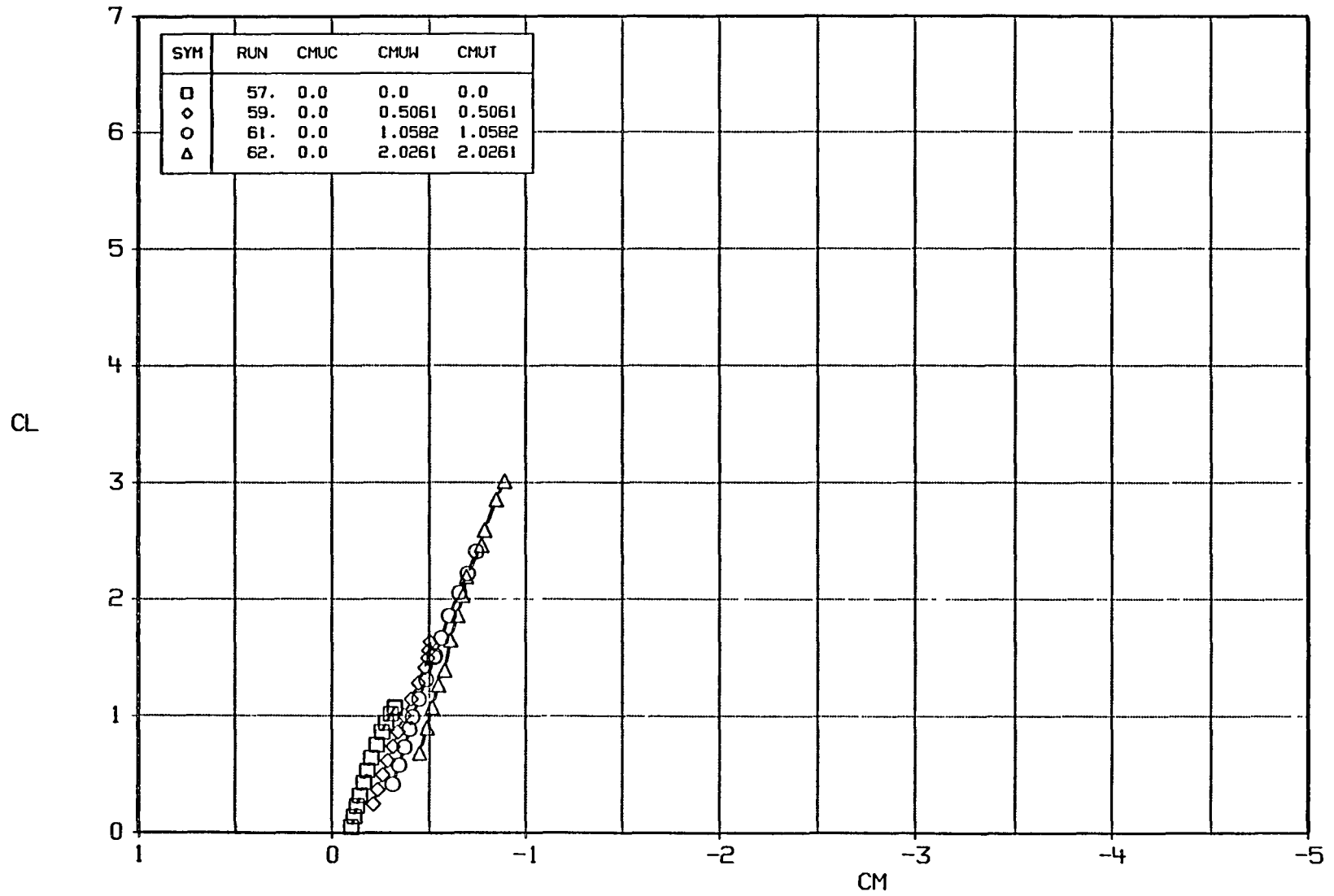


FIGURE A2c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=0, BN/B=1

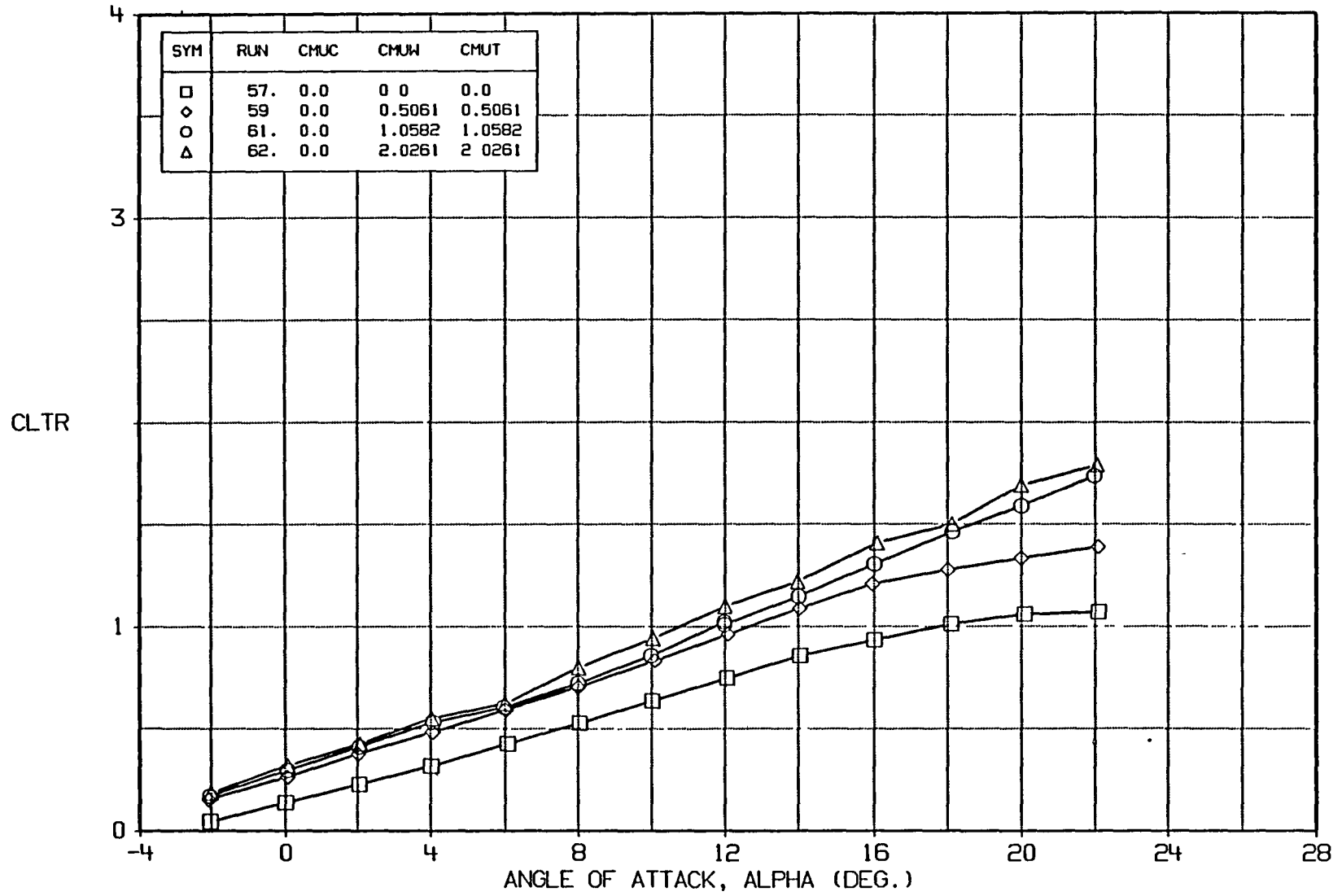


FIGURE A2d BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=0, BN/B=1

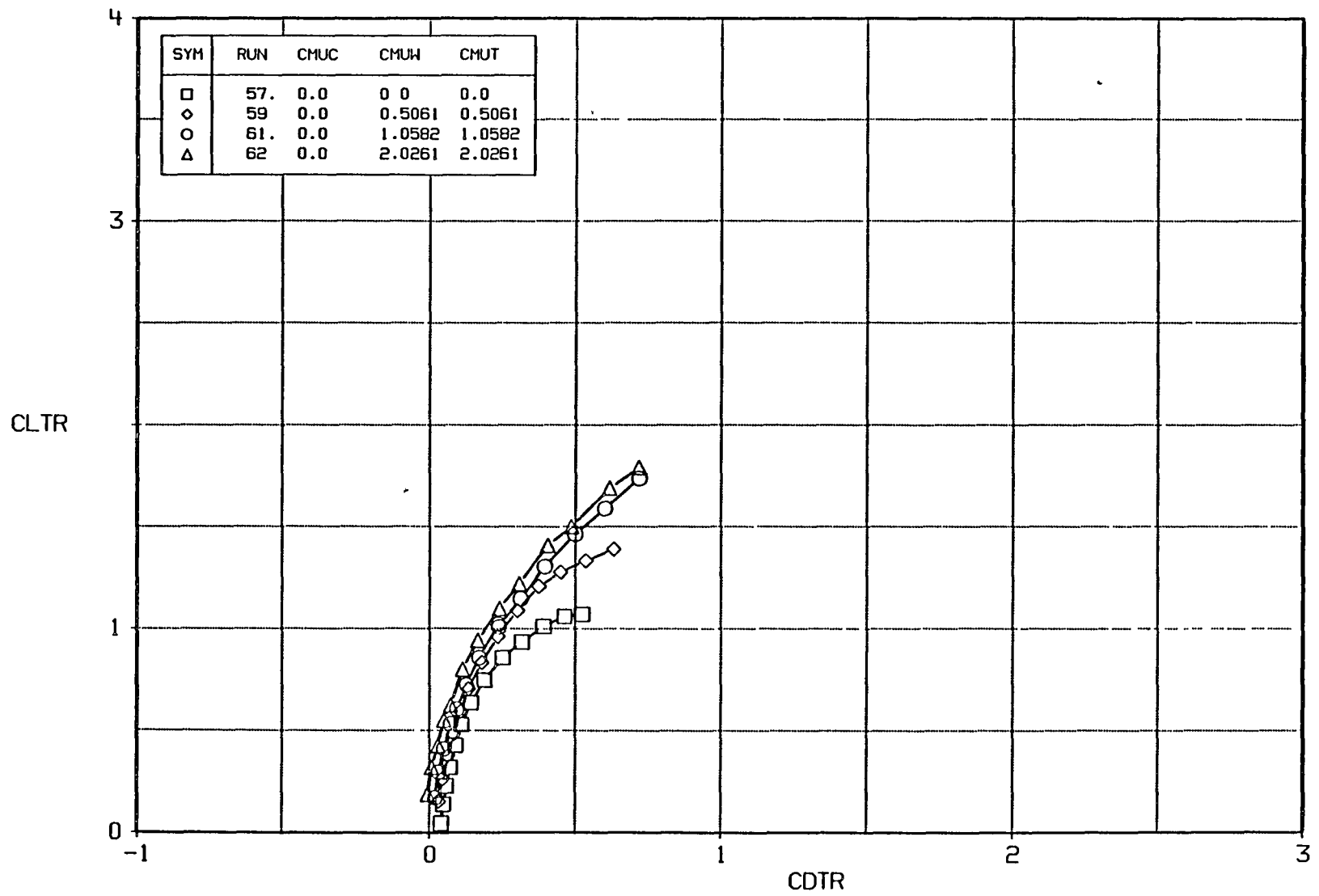


FIGURE A2e BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=0, BN/B=1

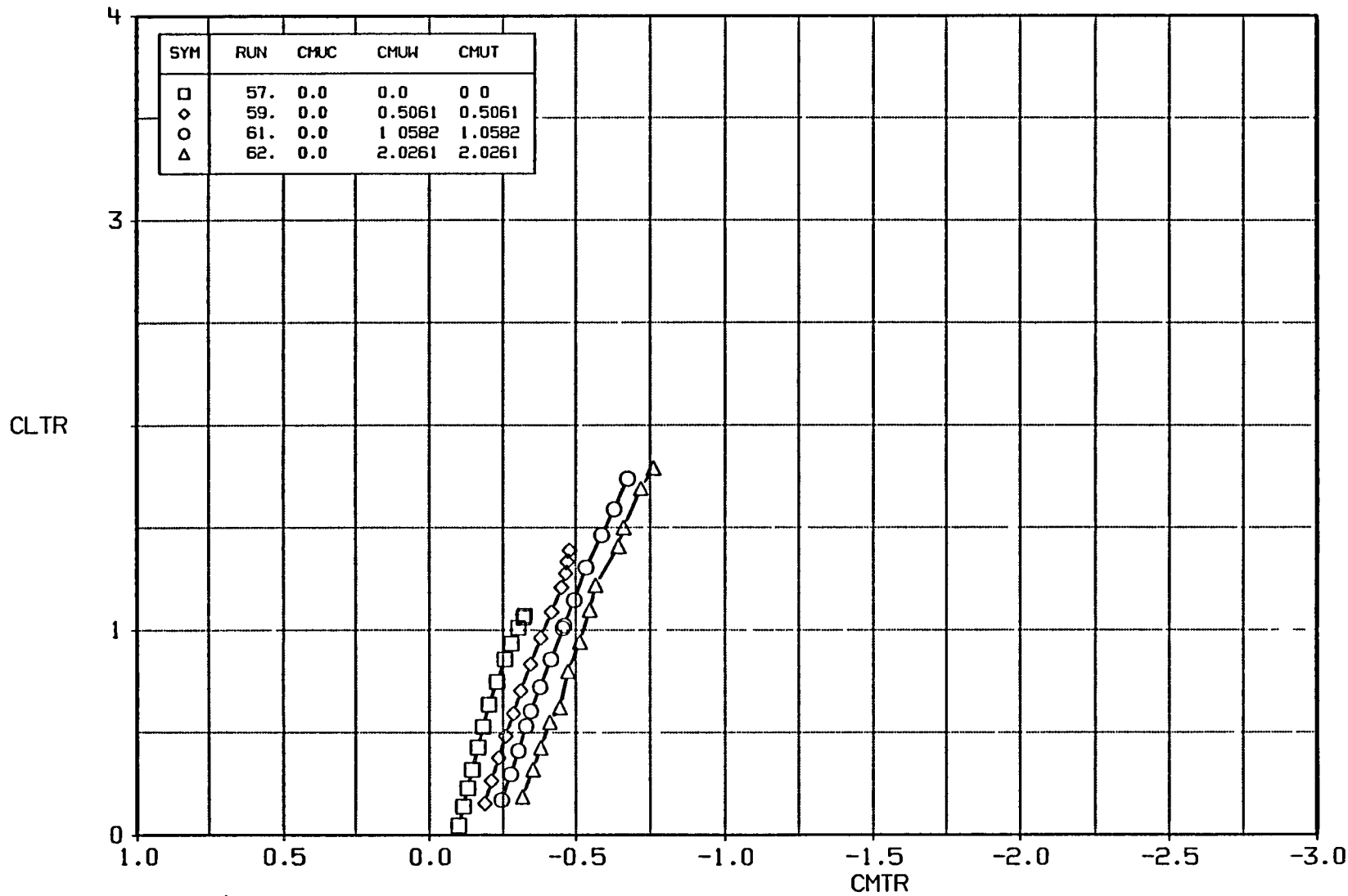


FIGURE A2f BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V, DELF=0, BN/B=1

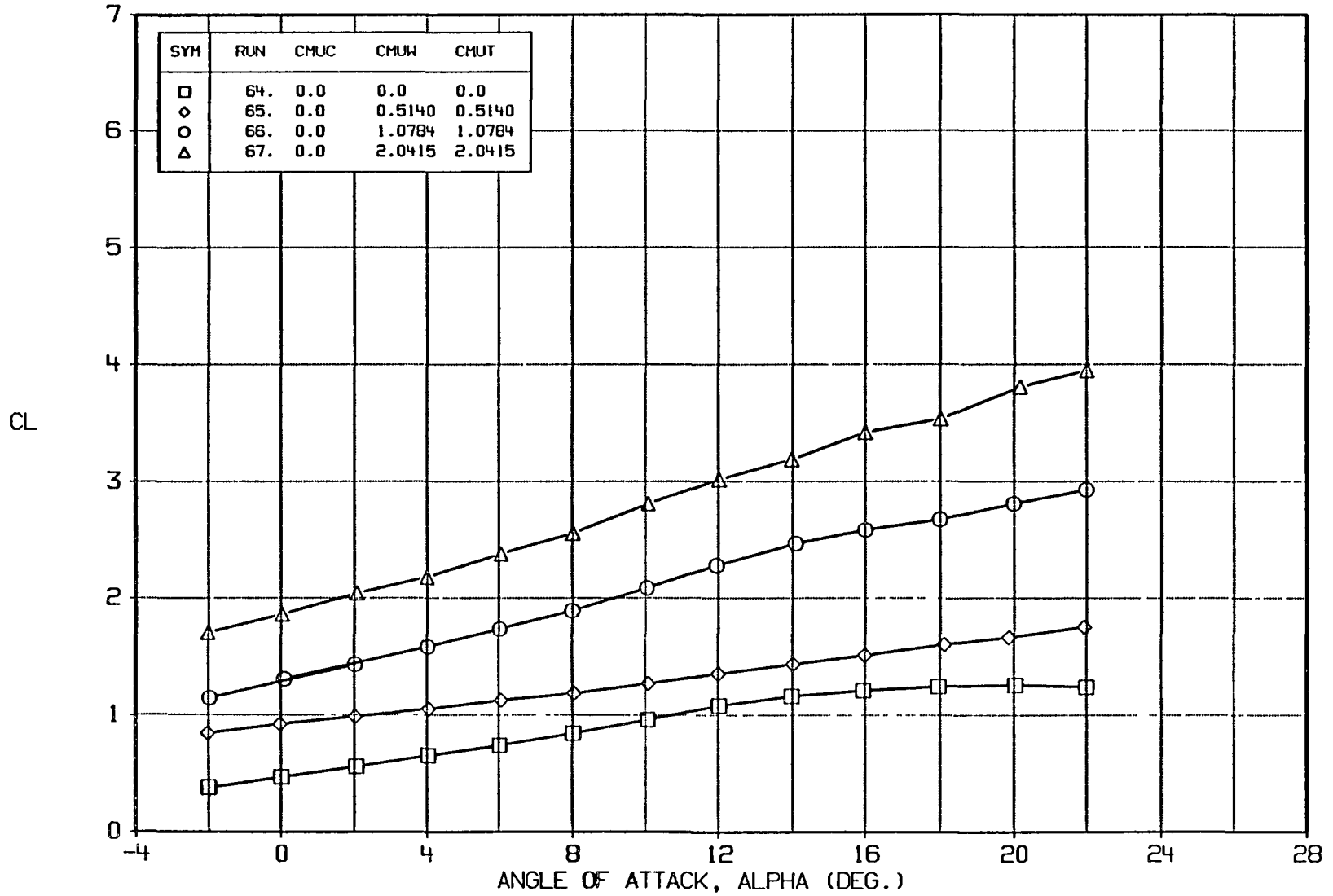


FIGURE A3a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=15, BN/B=1



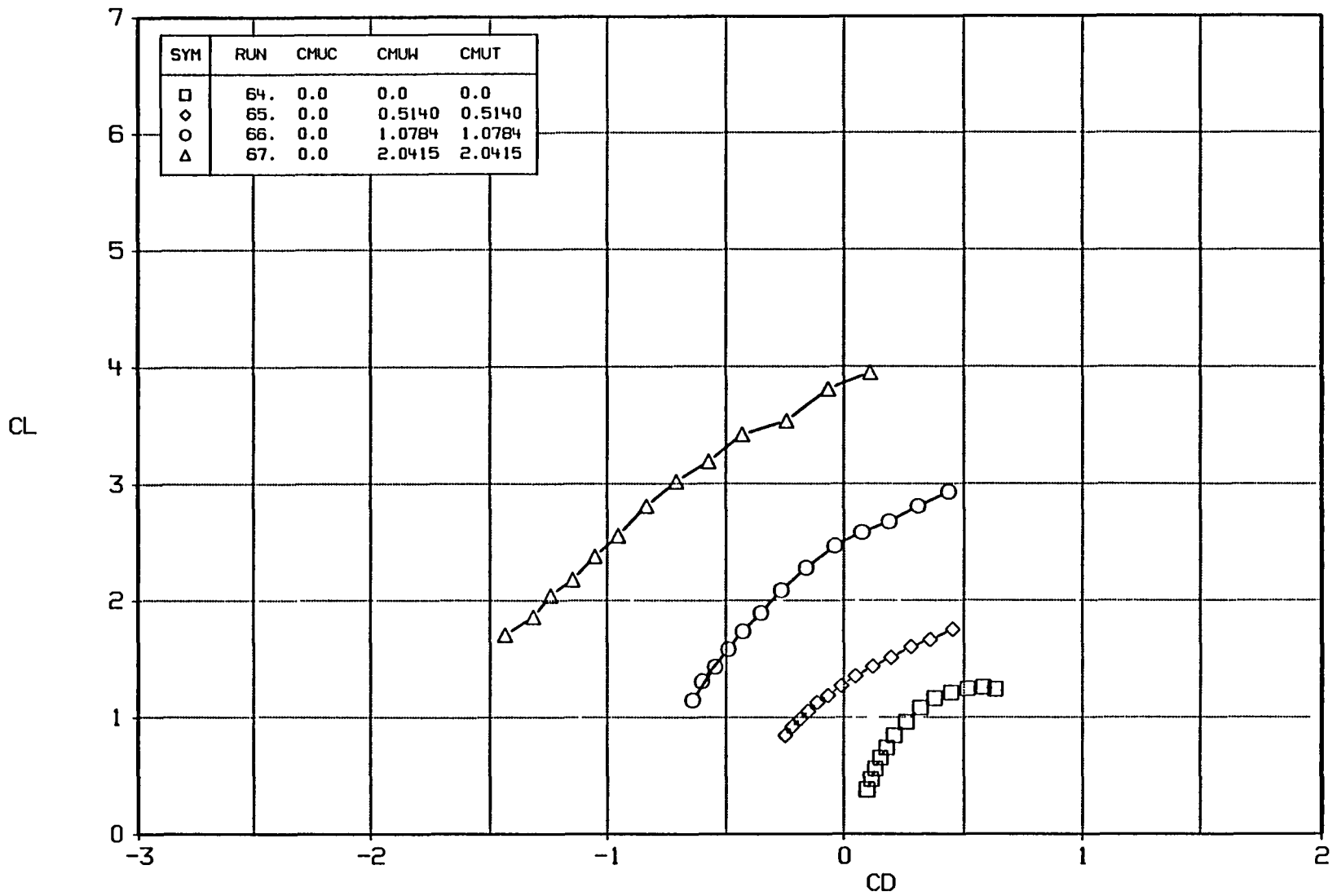


FIGURE A3b BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V, DELF=15, BN/B=1

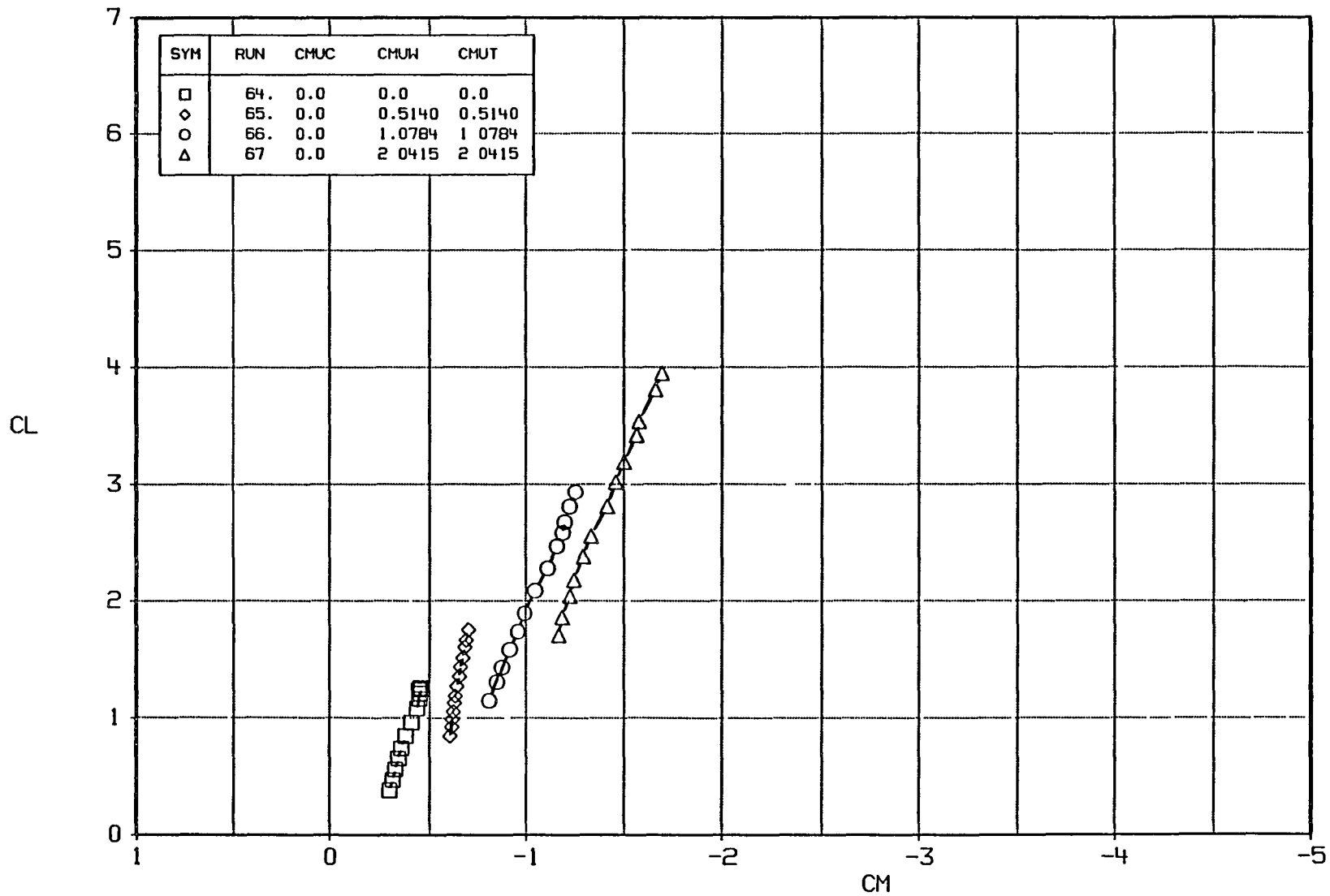


FIGURE A3c BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V, DELF=15, BN/B=1

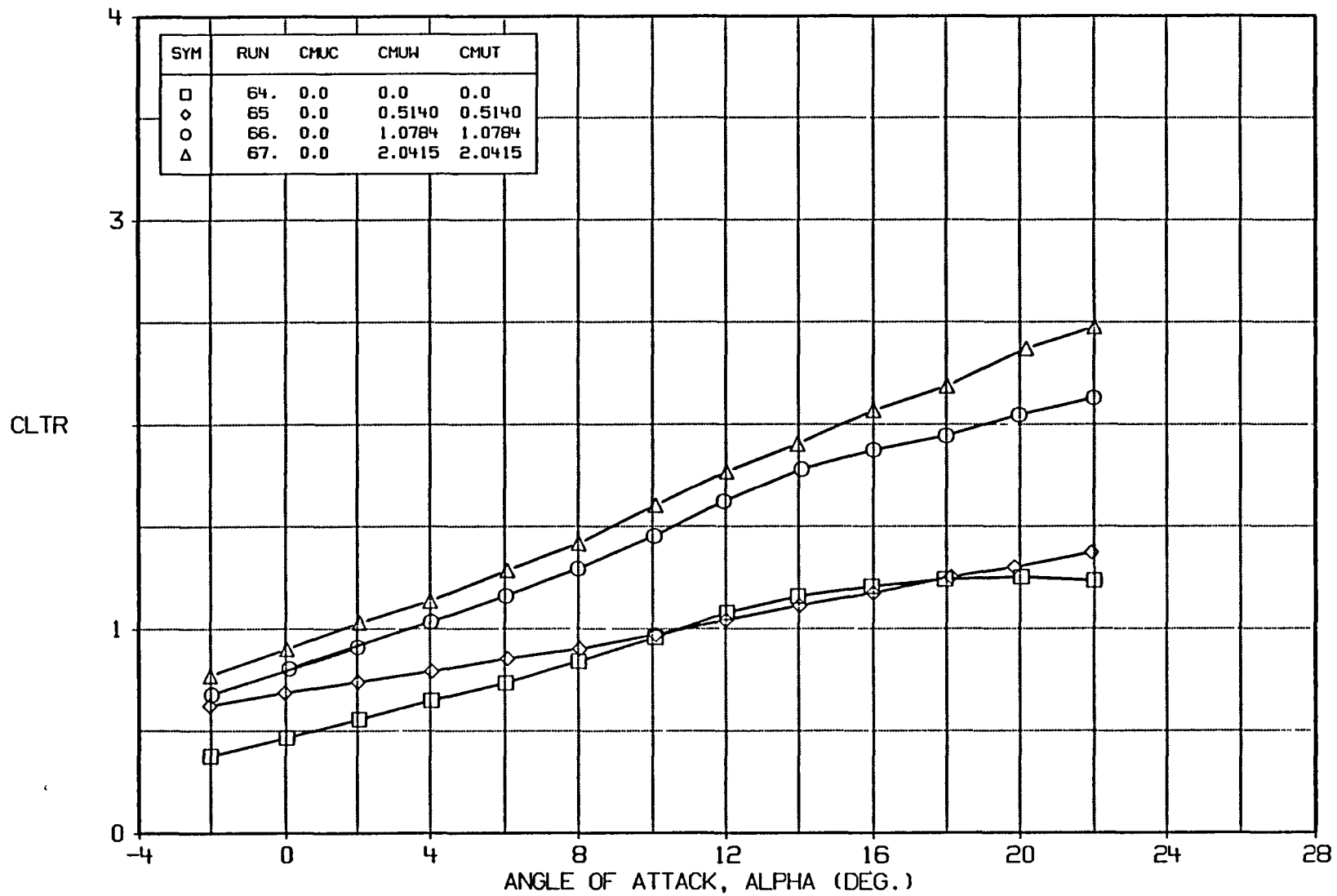


FIGURE A3d BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=15, BN/B=1

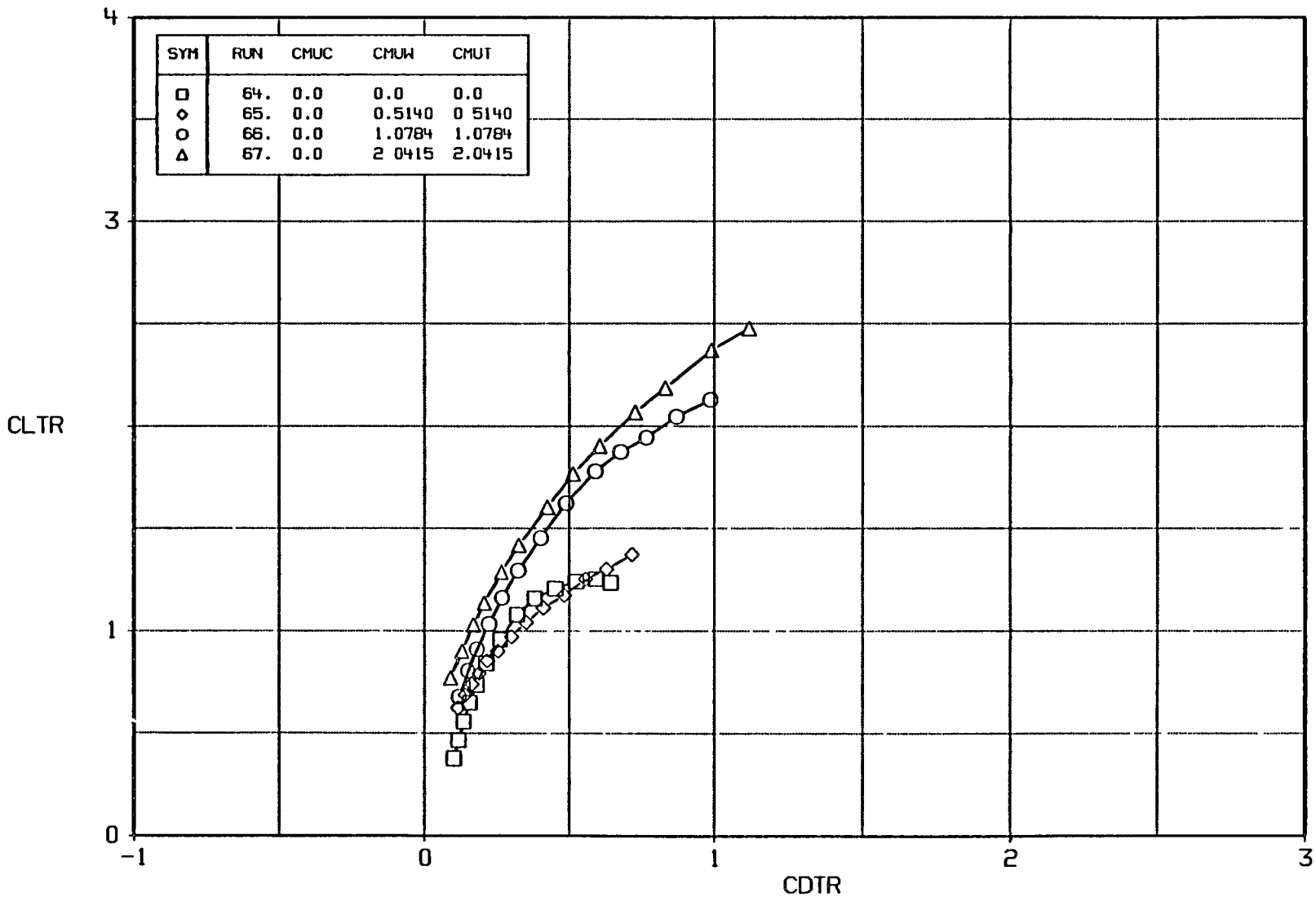


FIGURE A3e BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=15, BN/B=1

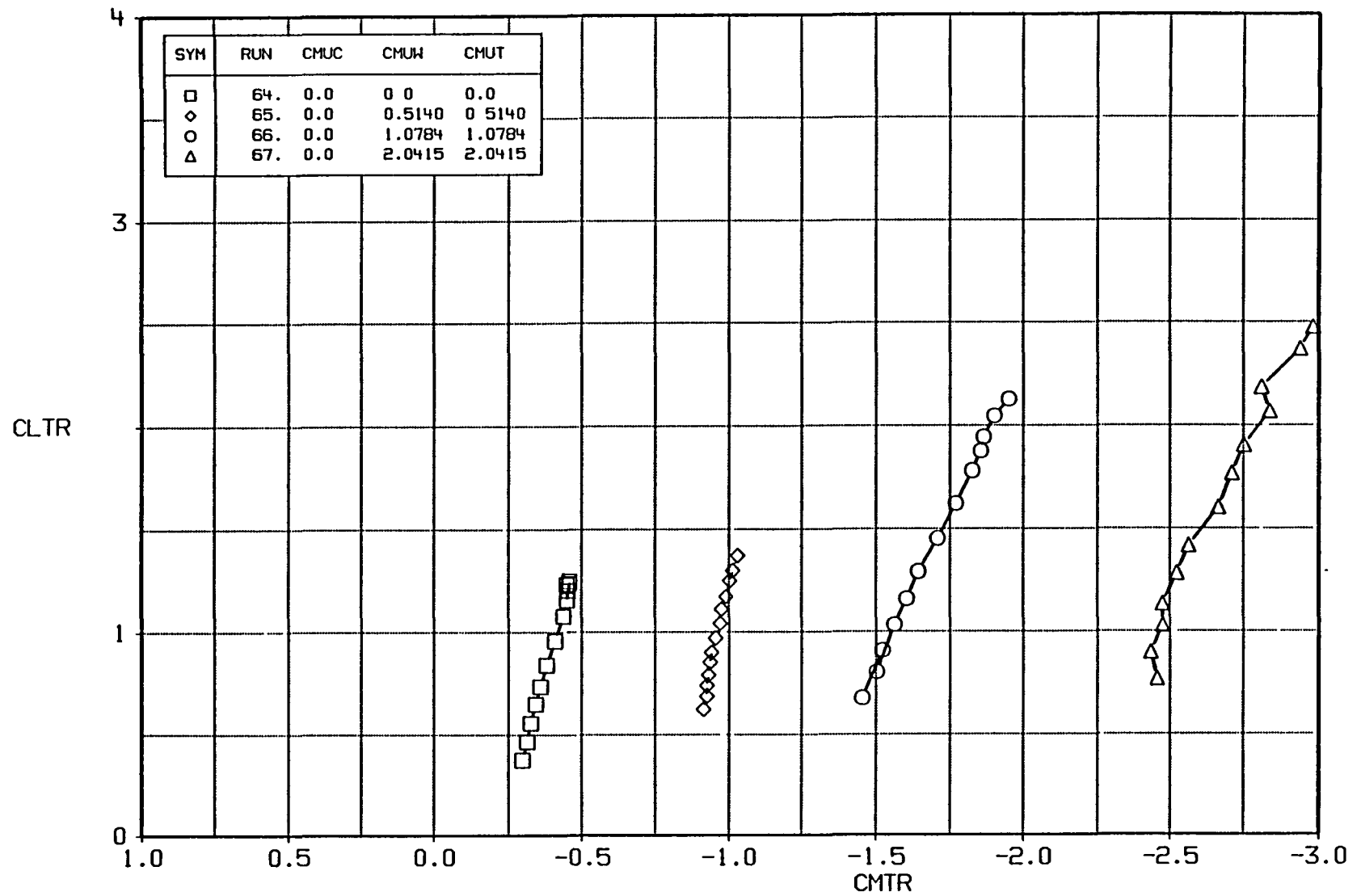


FIGURE A3f BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=15, BN/B=1

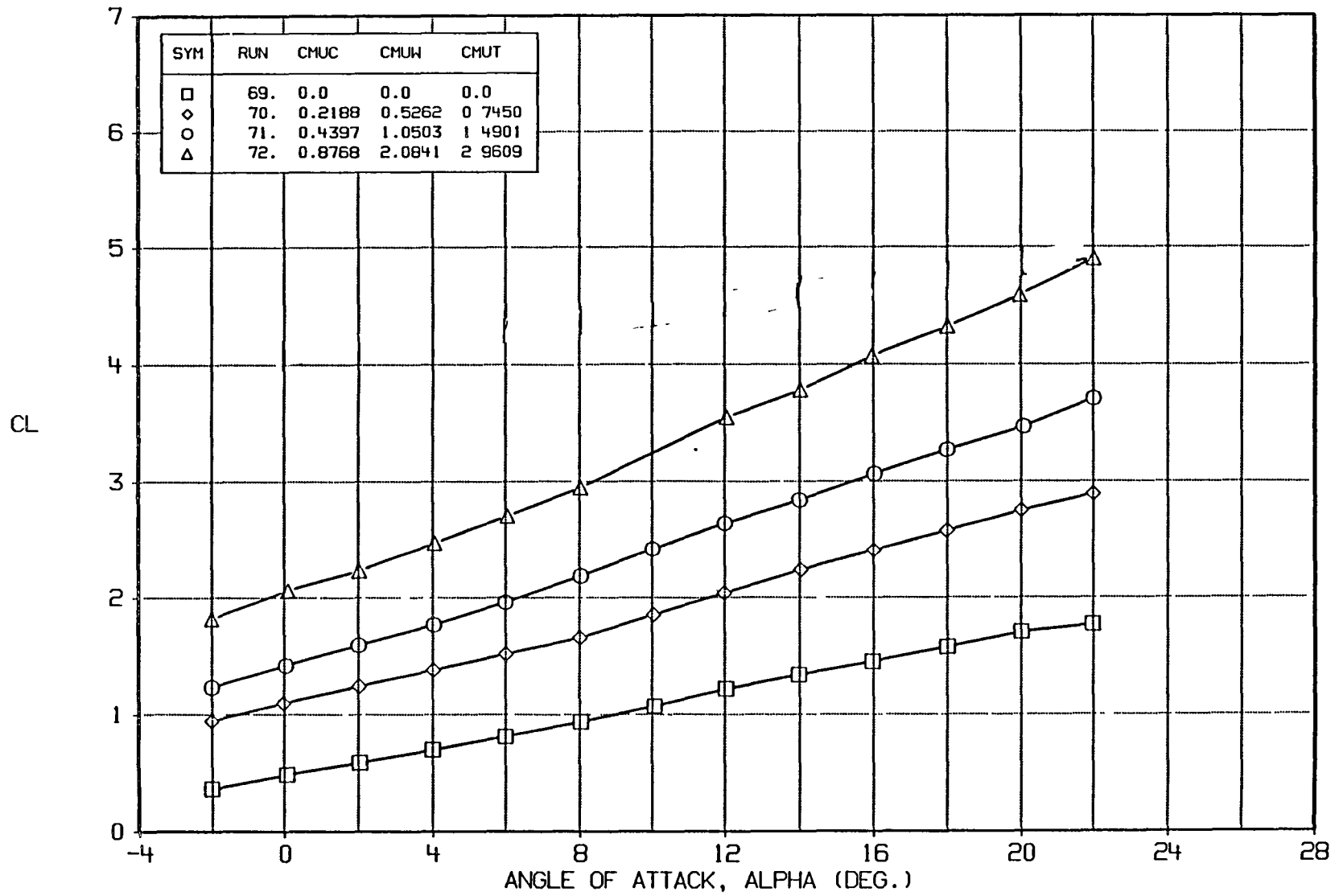


FIGURE A4a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=15, BN/B=1

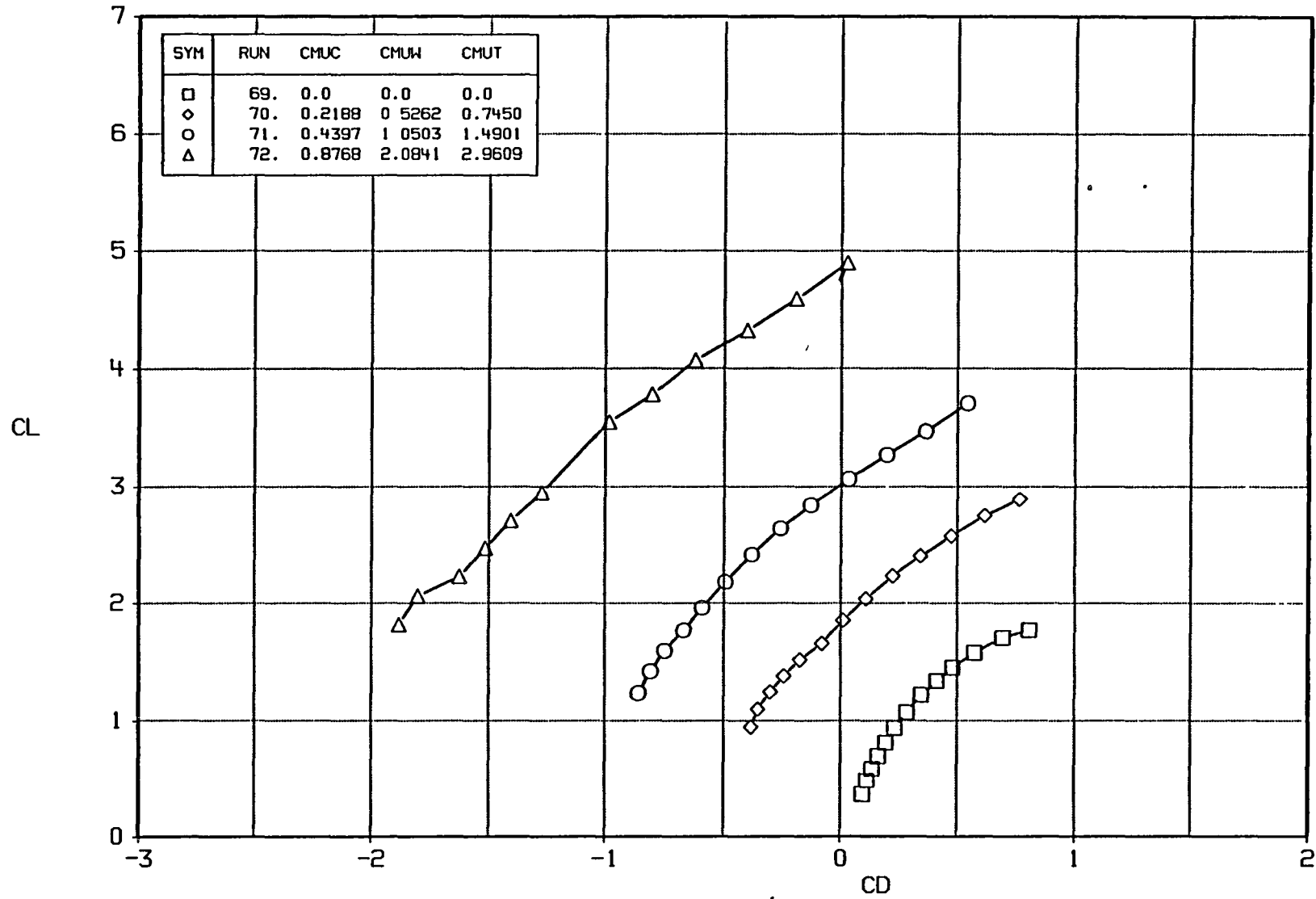


FIGURE A4b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=15, BN/B=1

A-30

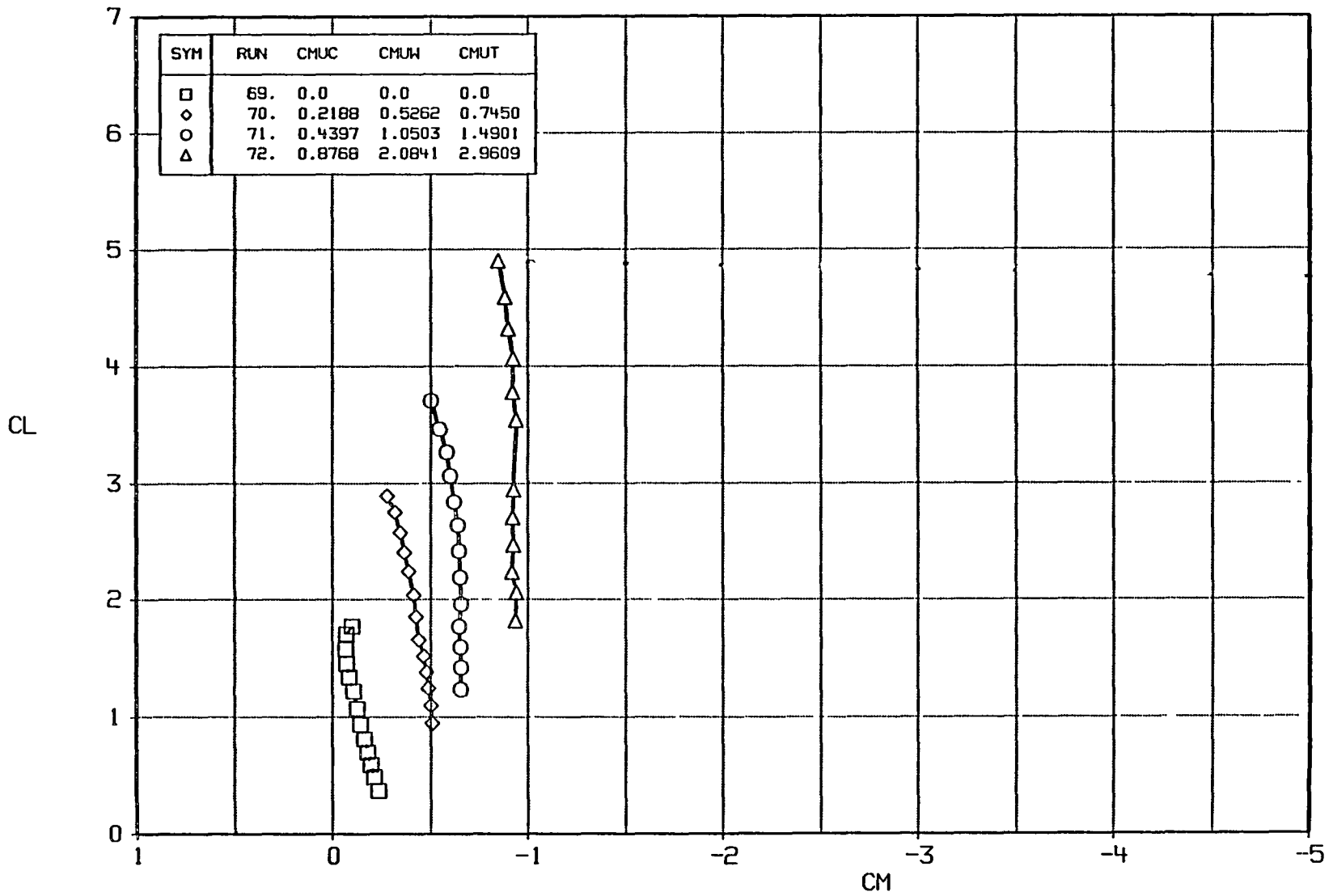


FIGURE A4c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=15, BN/B=1



A-31

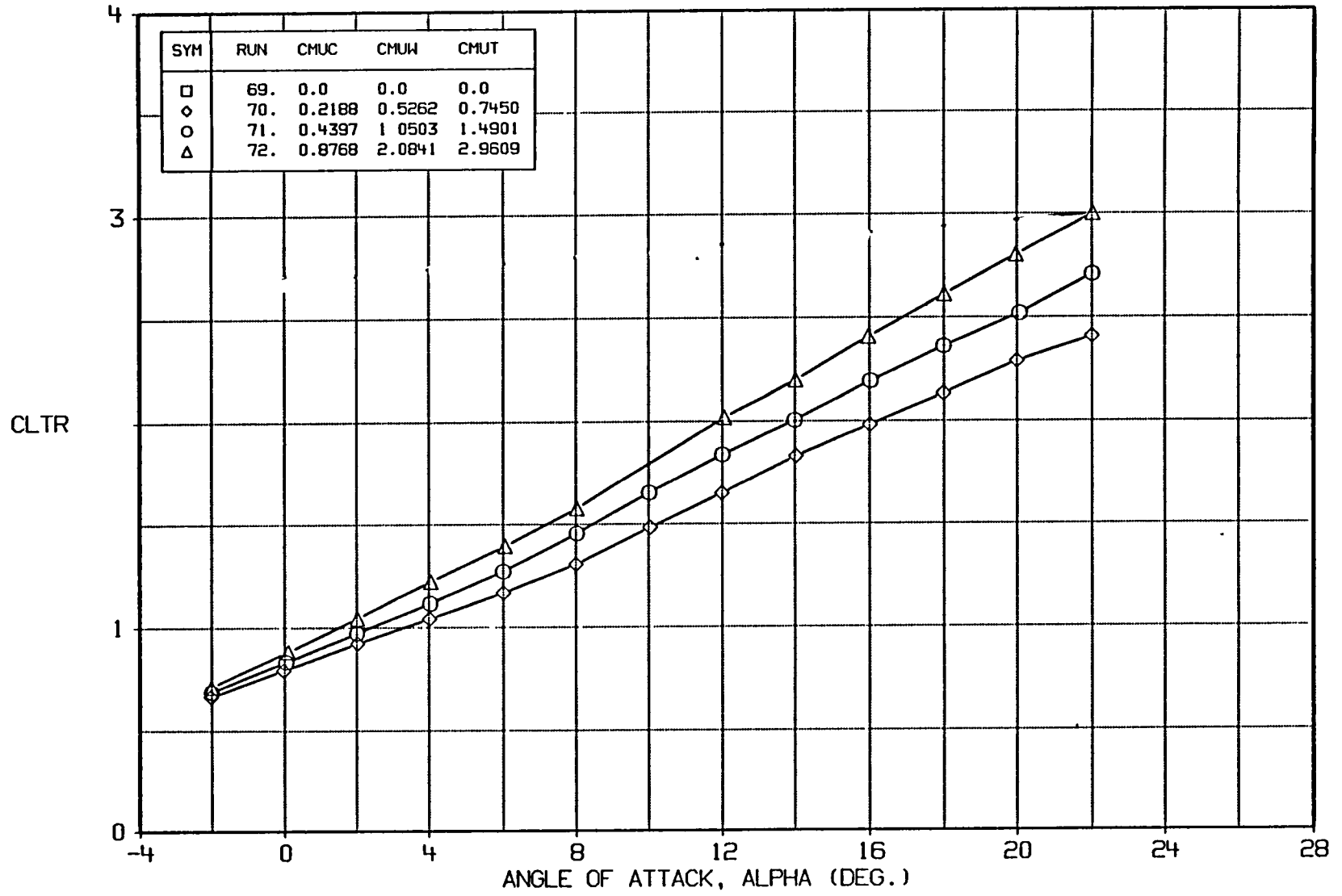


FIGURE A4d BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=15, BN/B=1

A-32

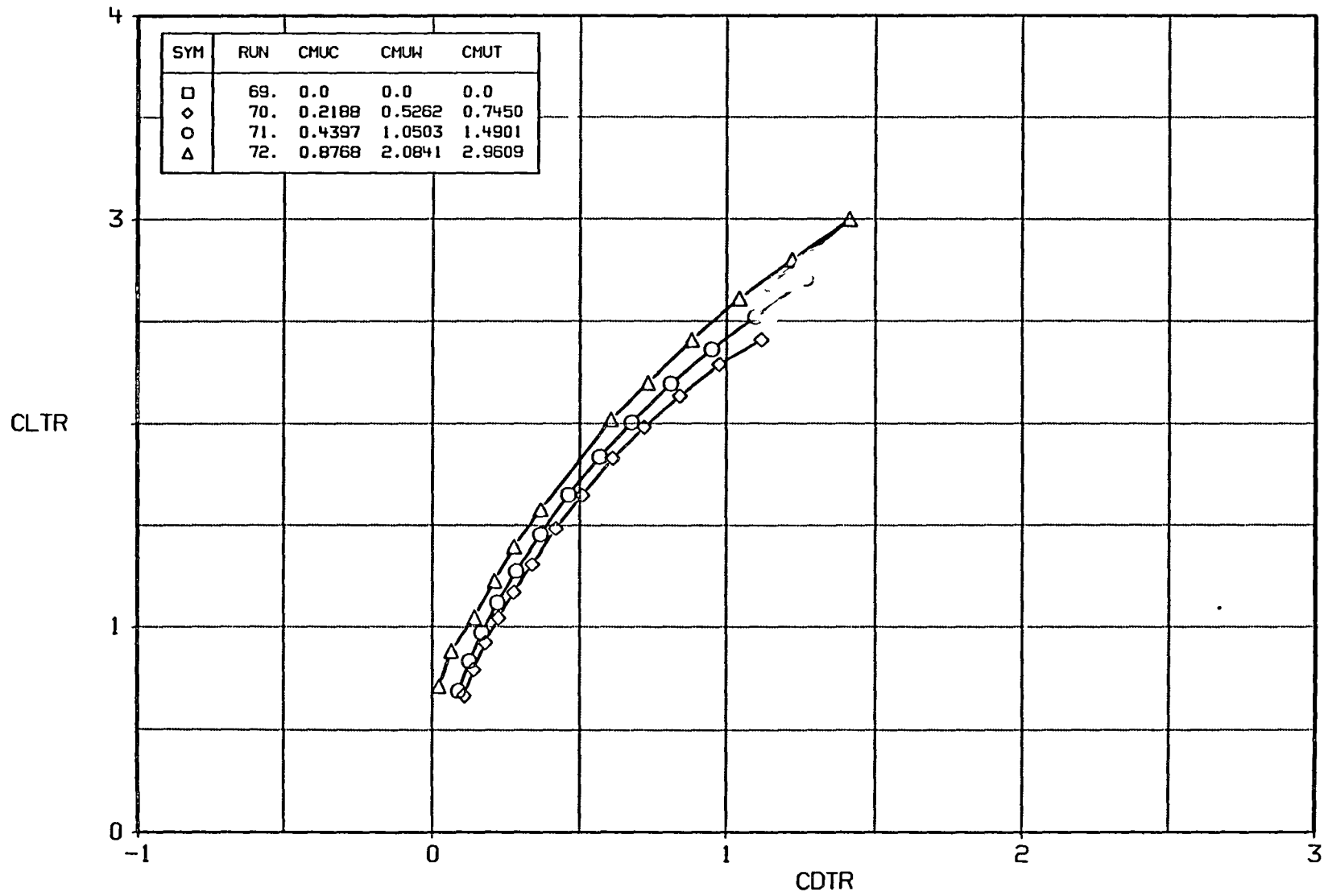


FIGURE A4e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=15, BN/B=1

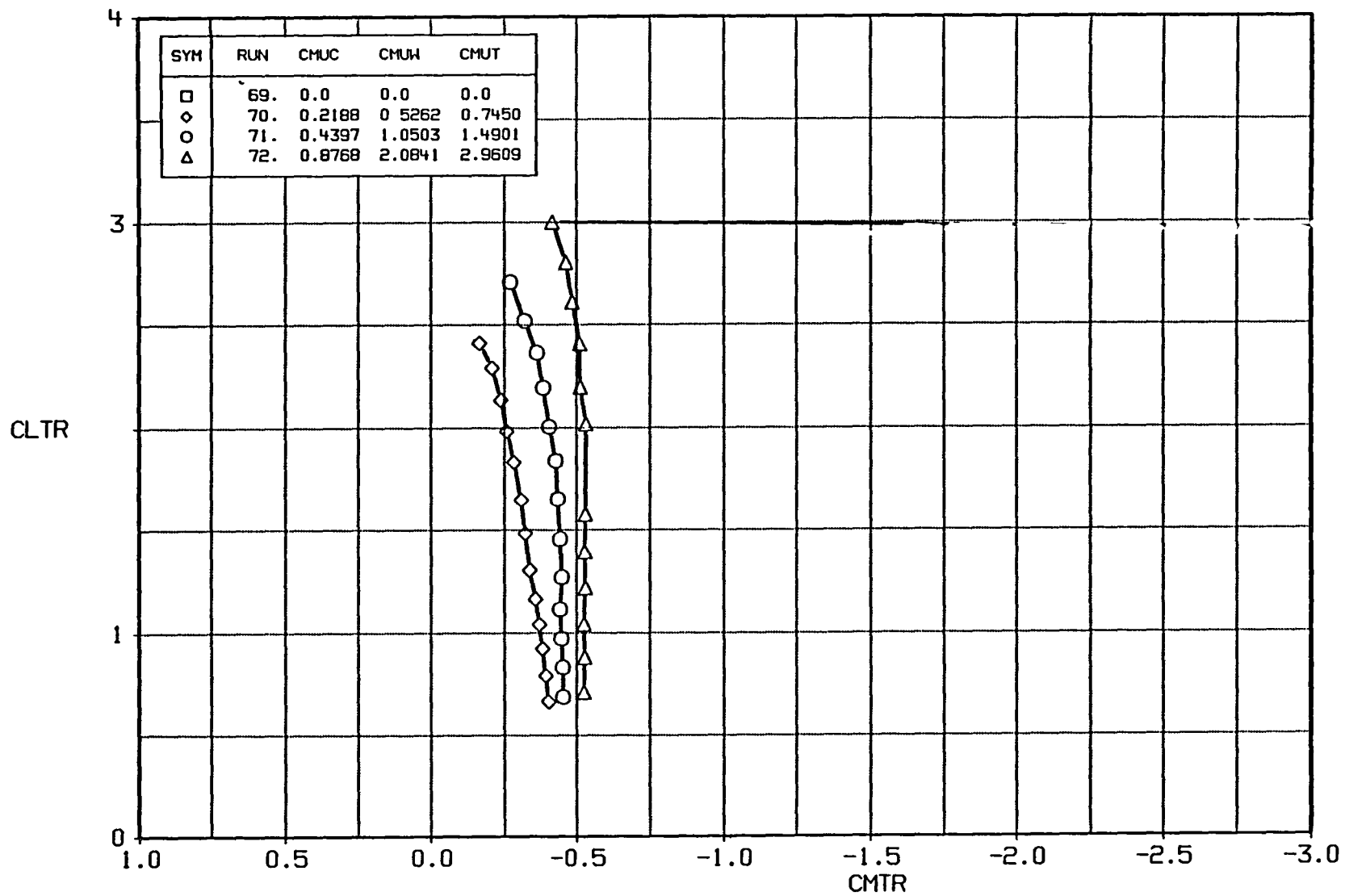


FIGURE A4f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=15, BN/B=1

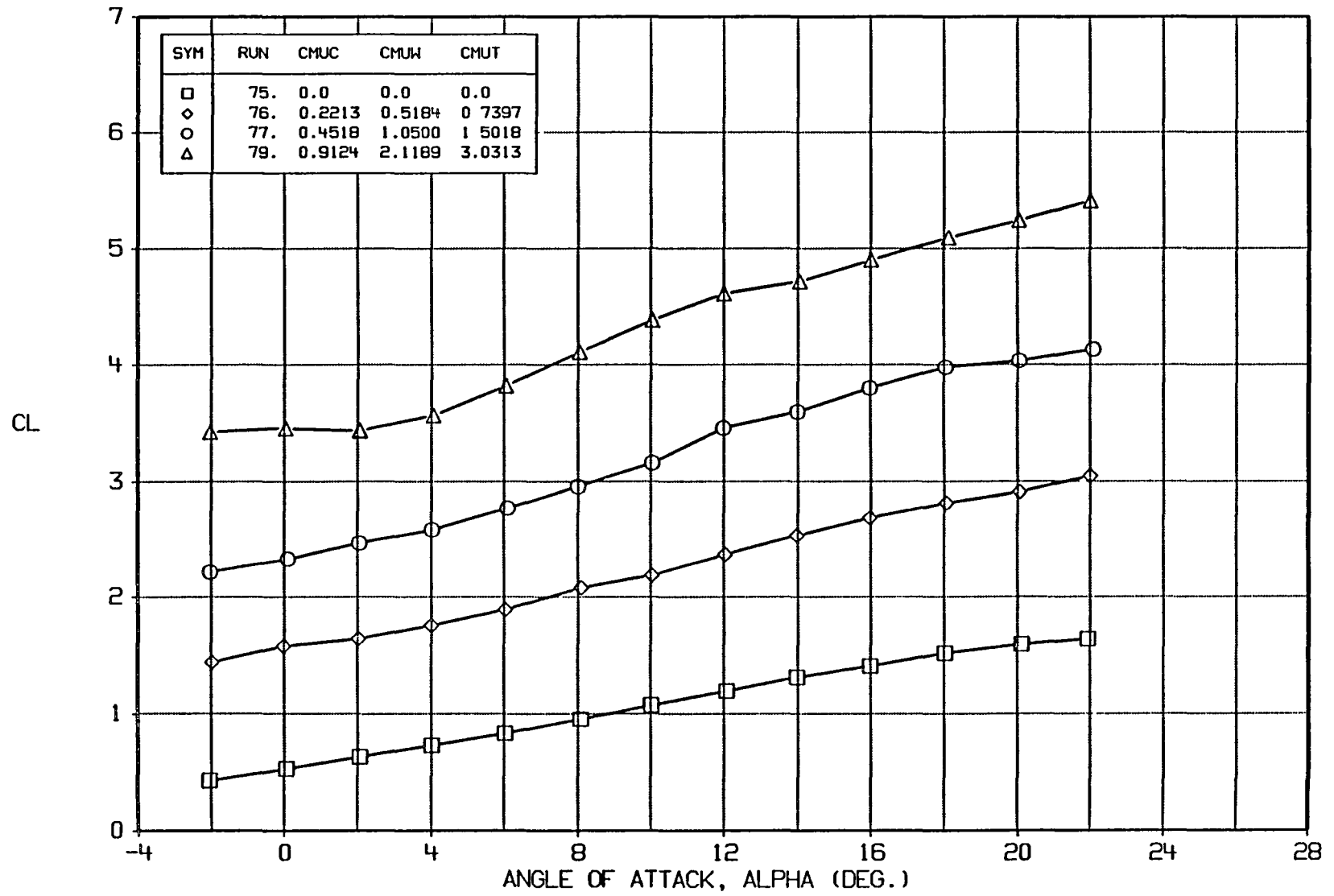


FIGURE A5a BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-35

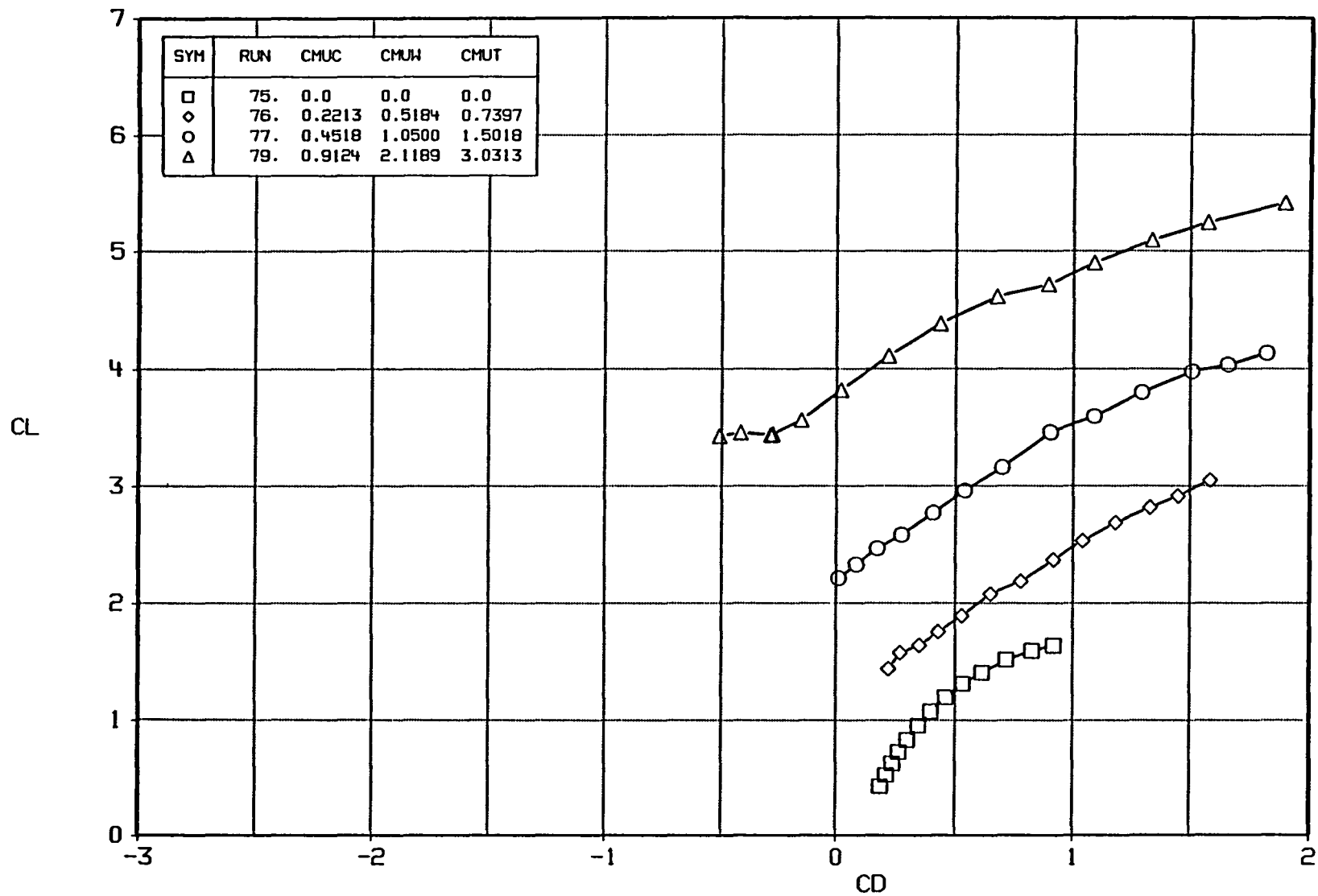


FIGURE A5b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

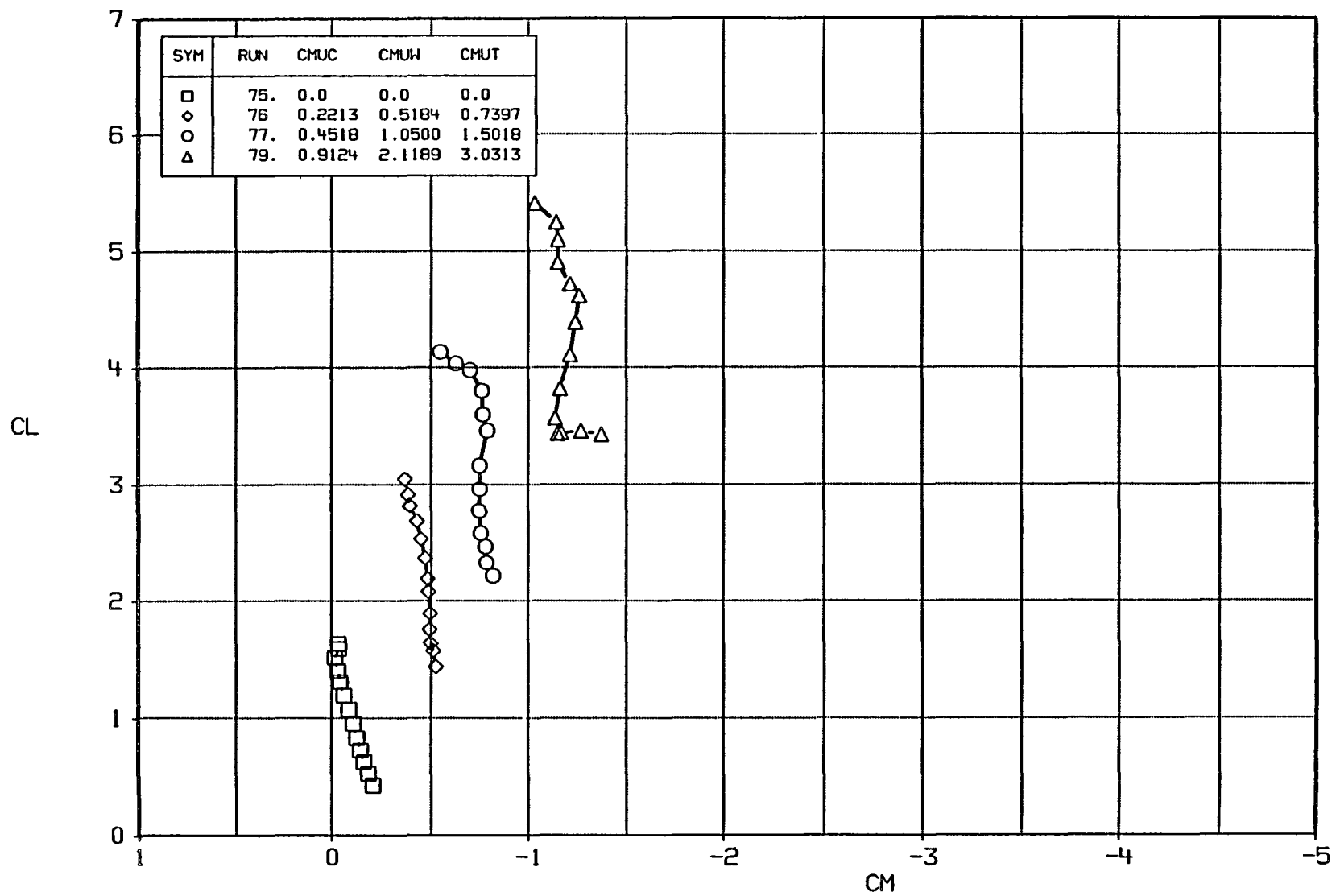


FIGURE A5c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

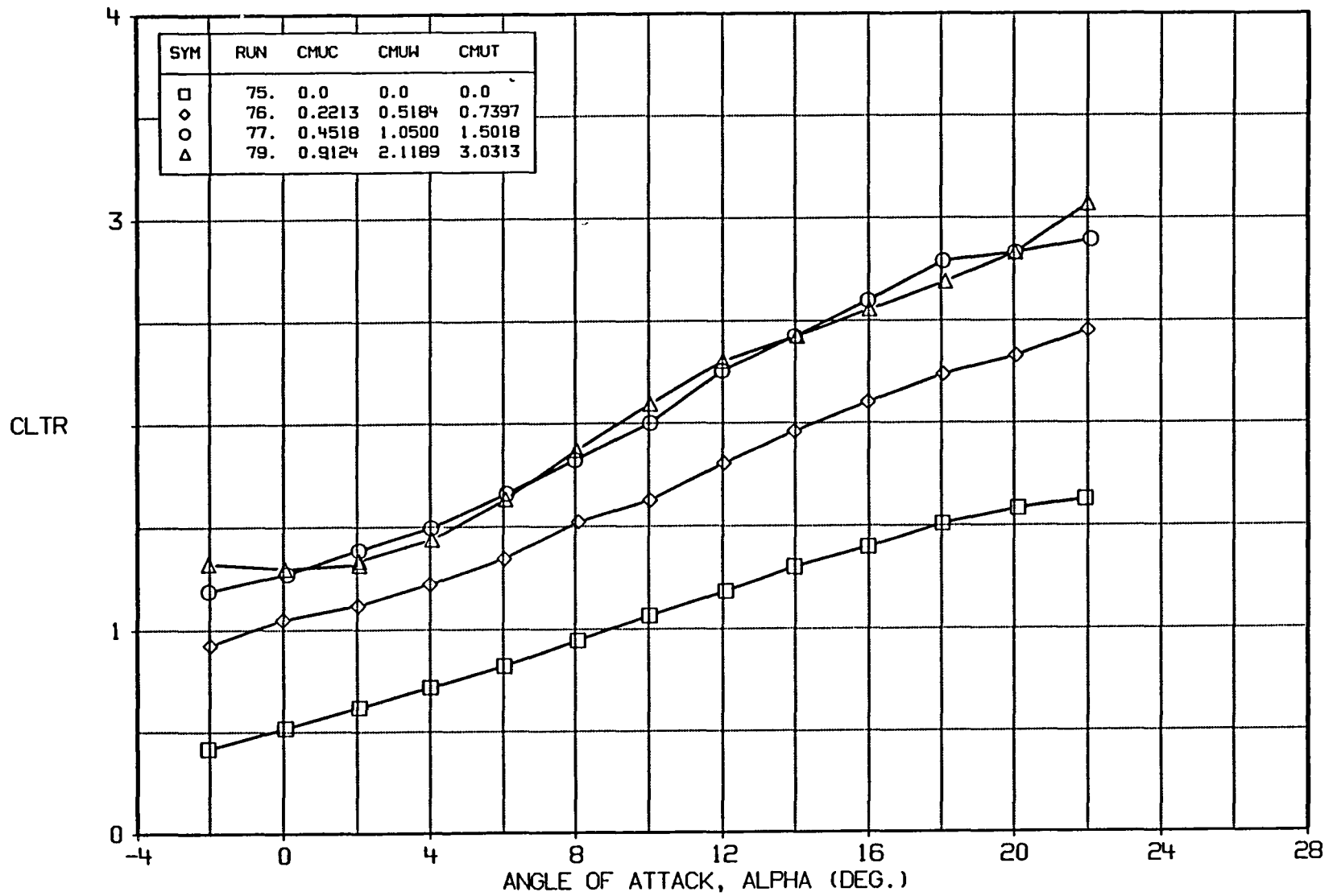


FIGURE A5d BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V, DELF=45, BN/B=1

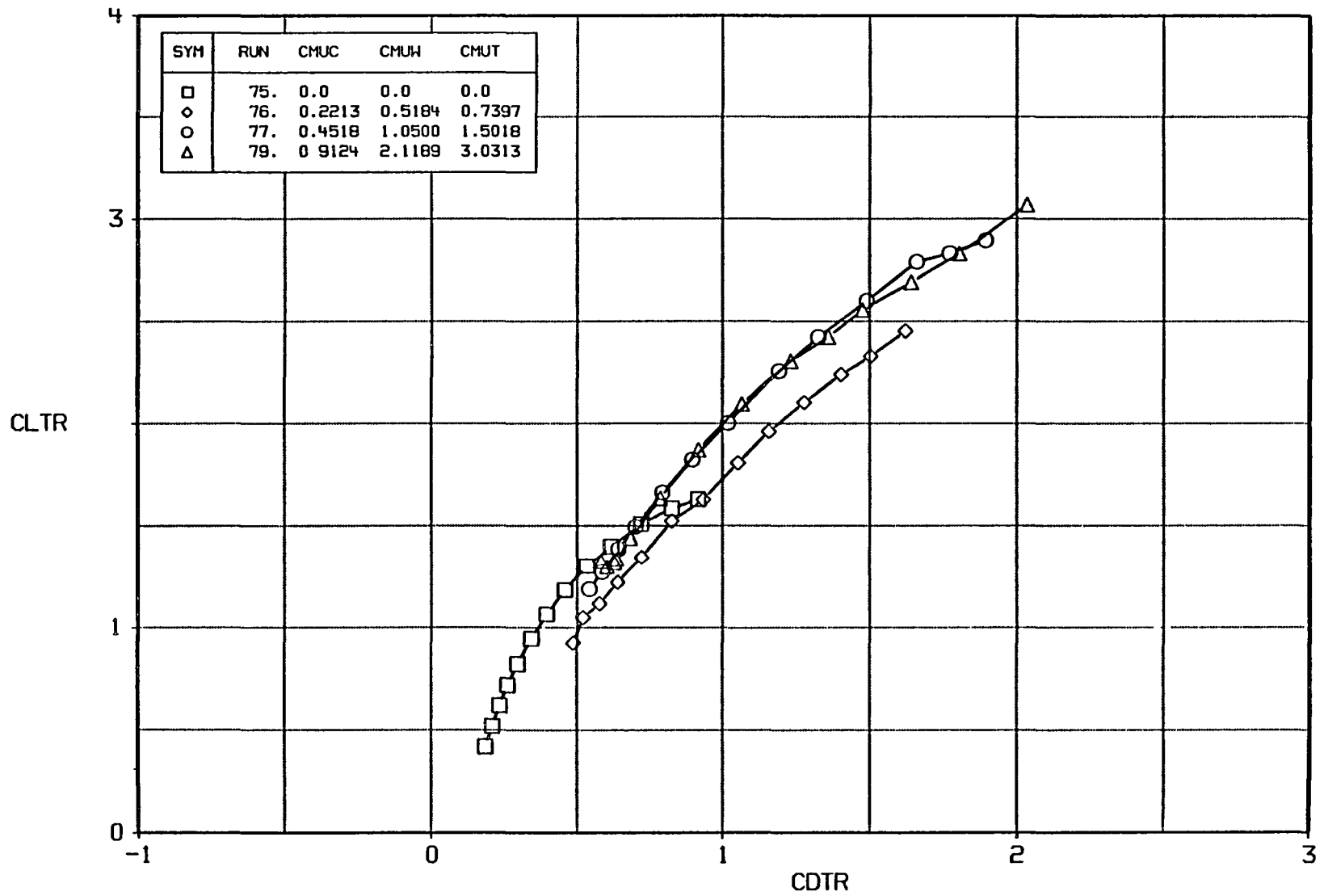


FIGURE A5e BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V, DELF=45, BN/B=1



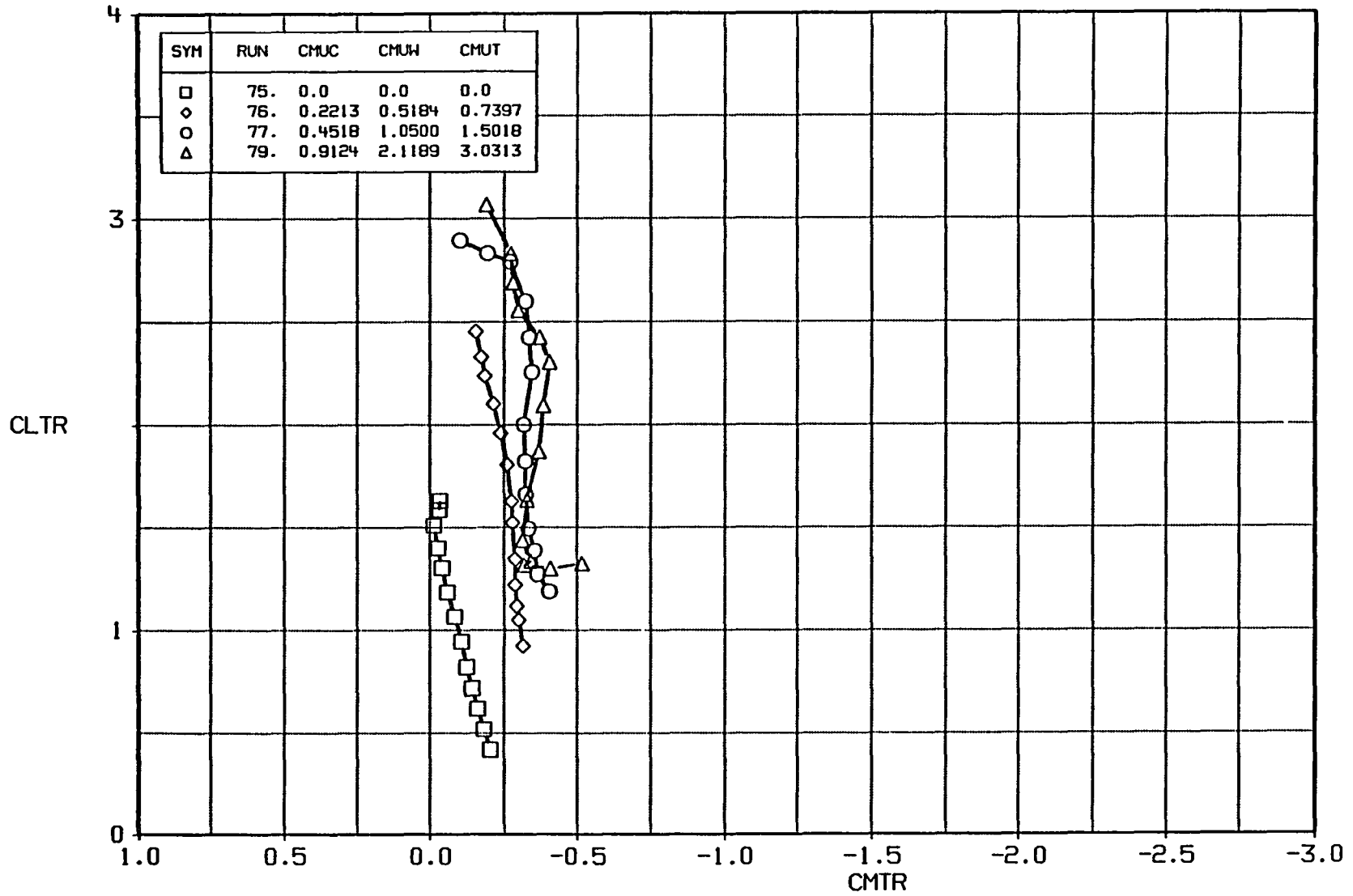


FIGURE A5f BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-40

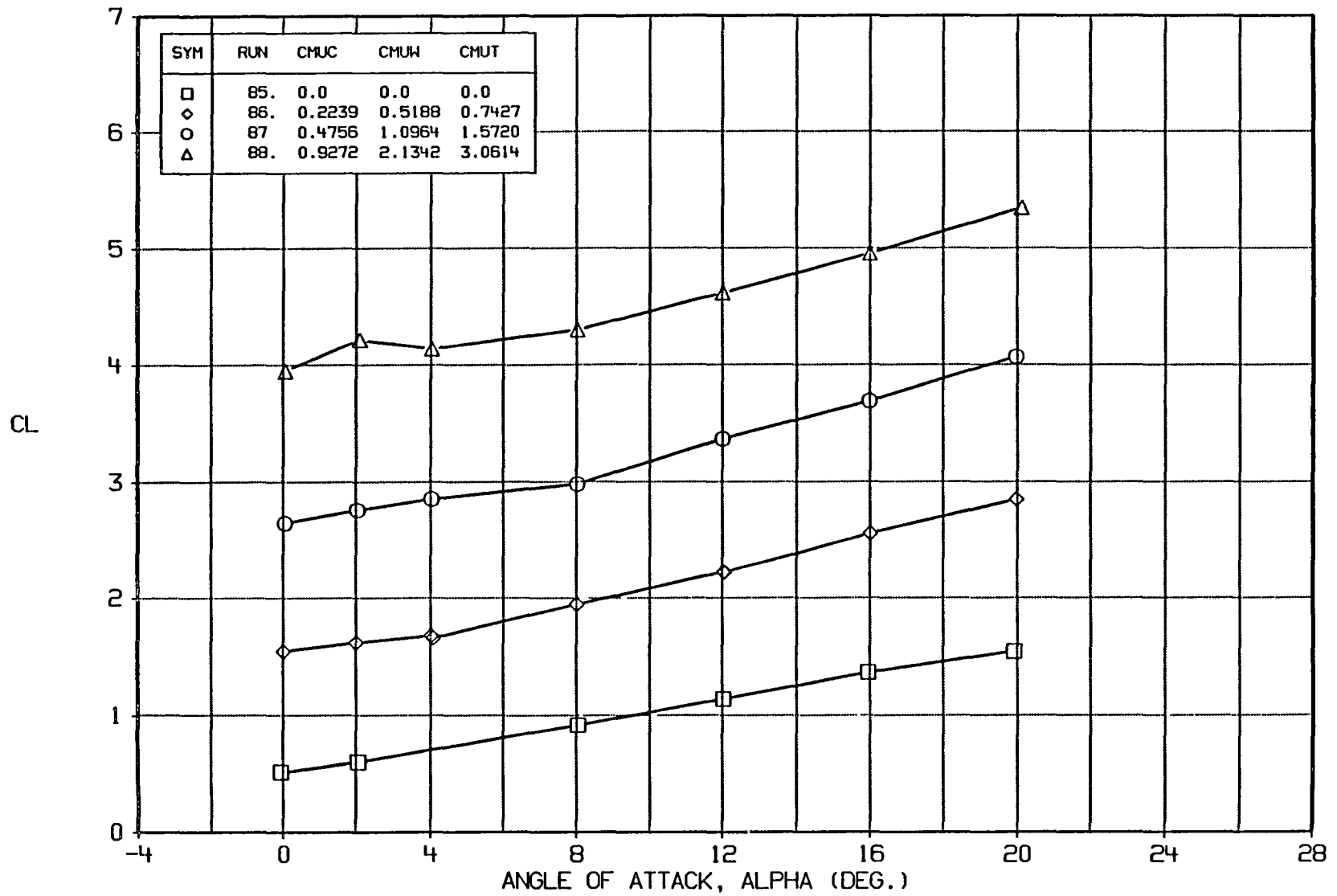


FIGURE A6a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W6V, DELF=45, BN/B=1

A-41

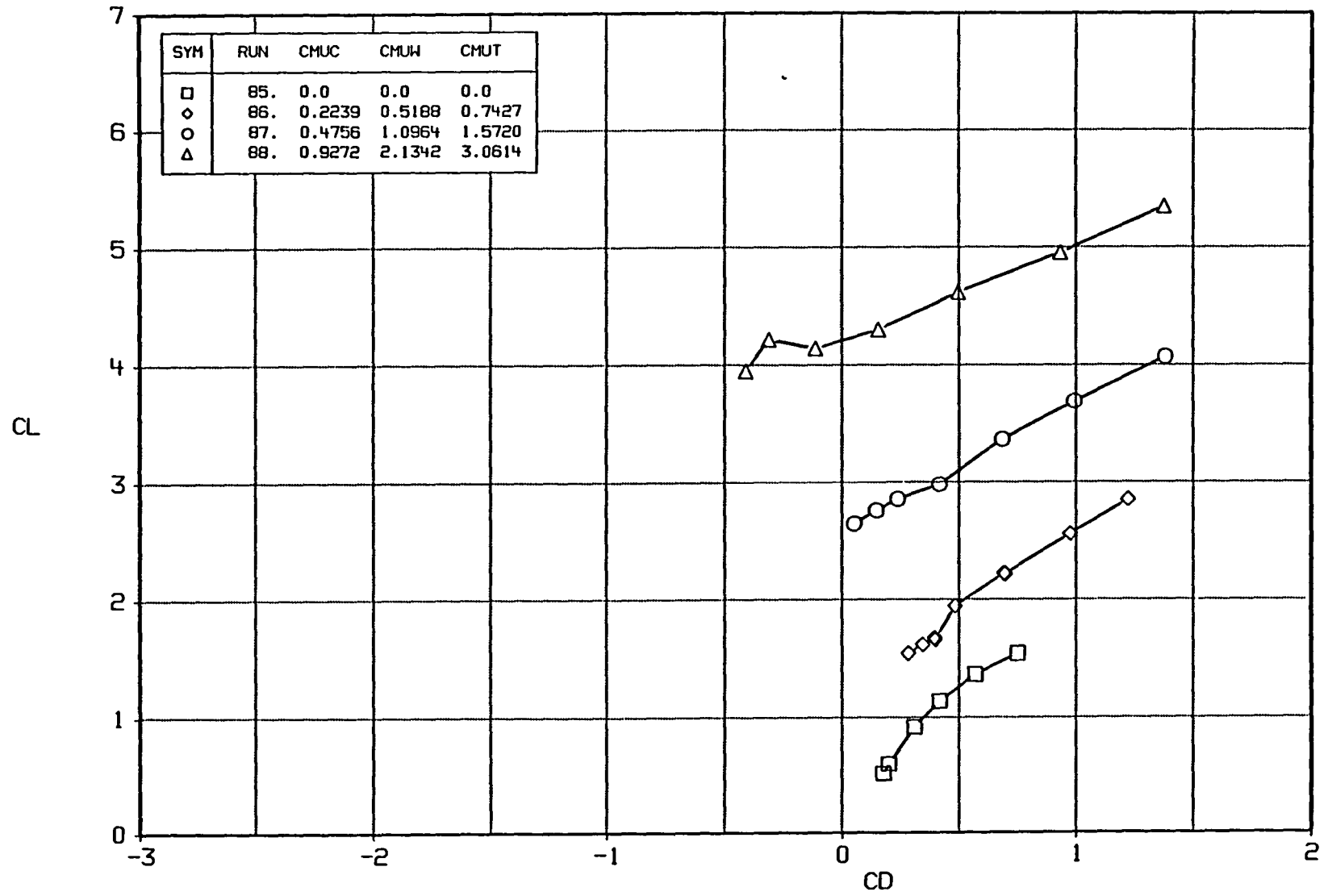


FIGURE A6b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W6V, DELF=45, BN/B=1

A-42

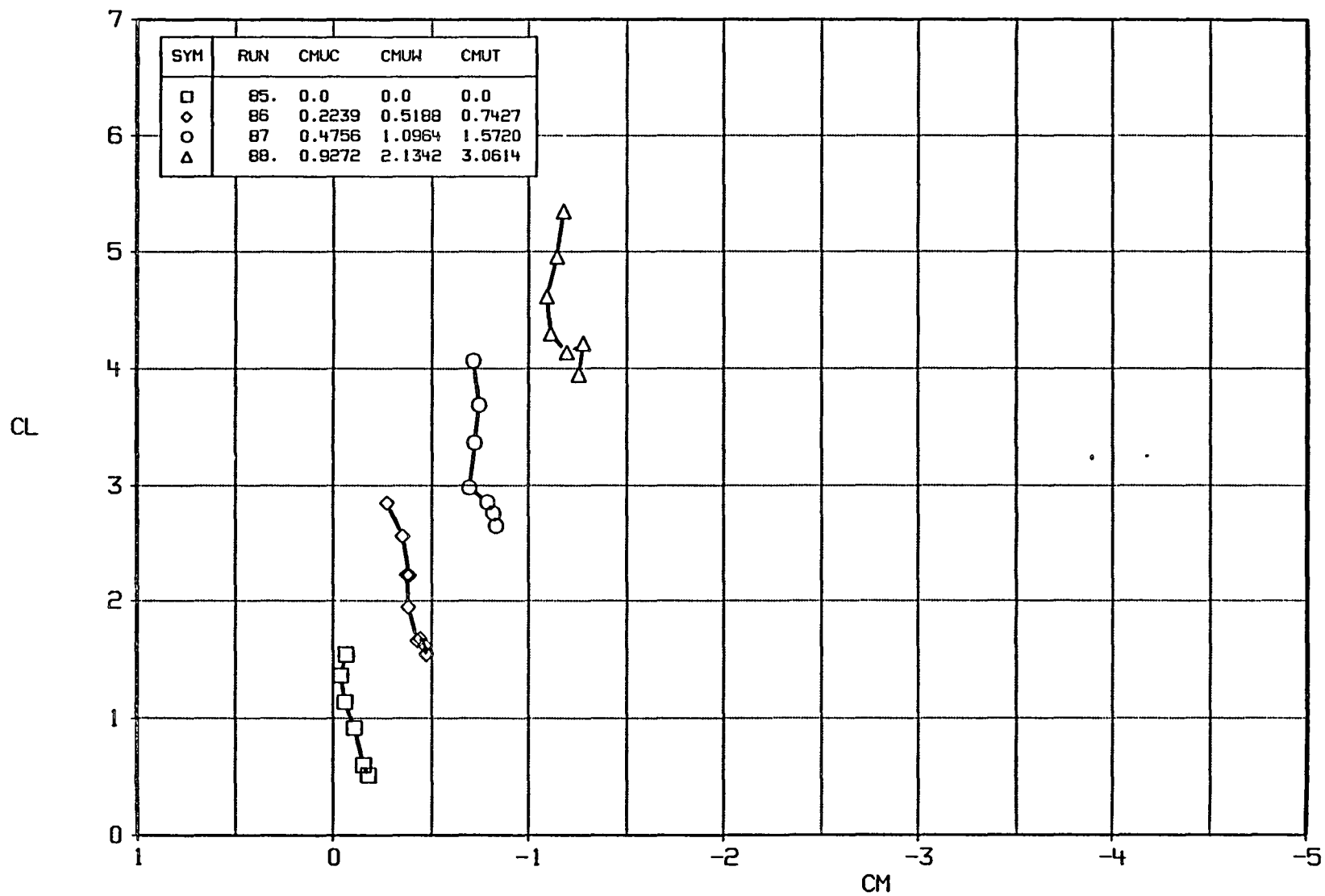


FIGURE A6c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W6V, DELF=45, BN/B=1

A-43

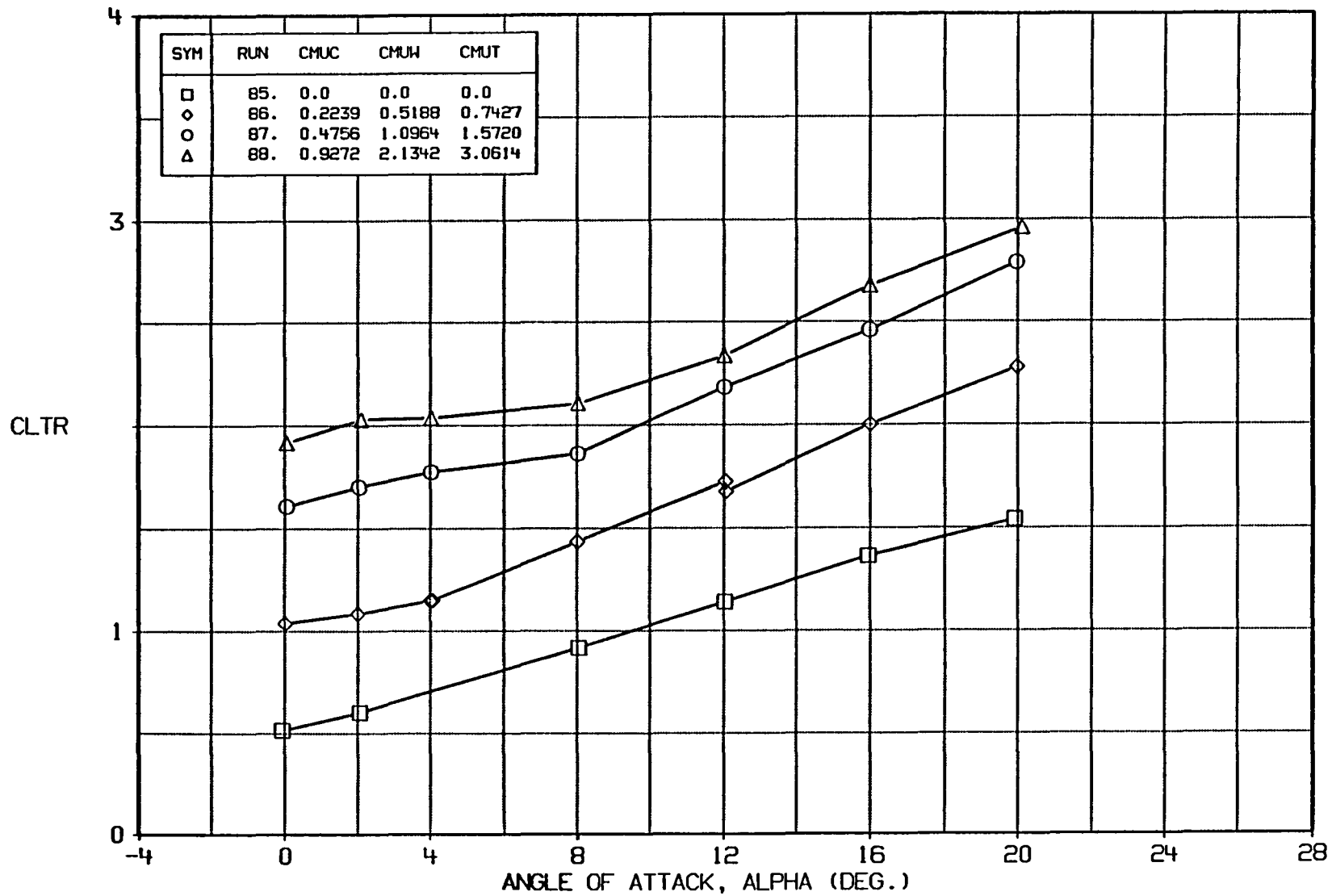


FIGURE A6d BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W6V, DELF=45, BN/B=1

A-44

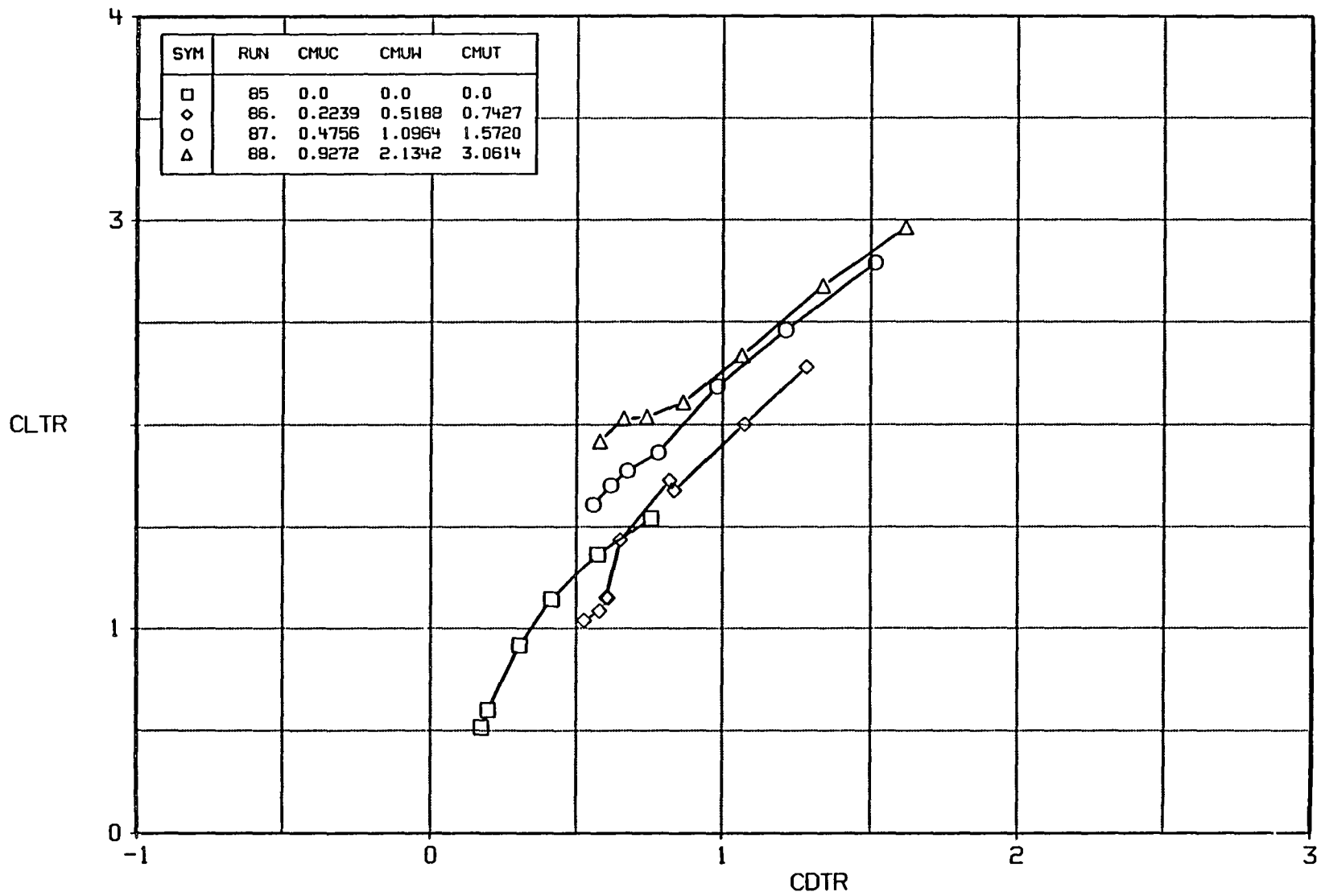


FIGURE A6e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W6V, DELF=45, BN/B=1

A-45

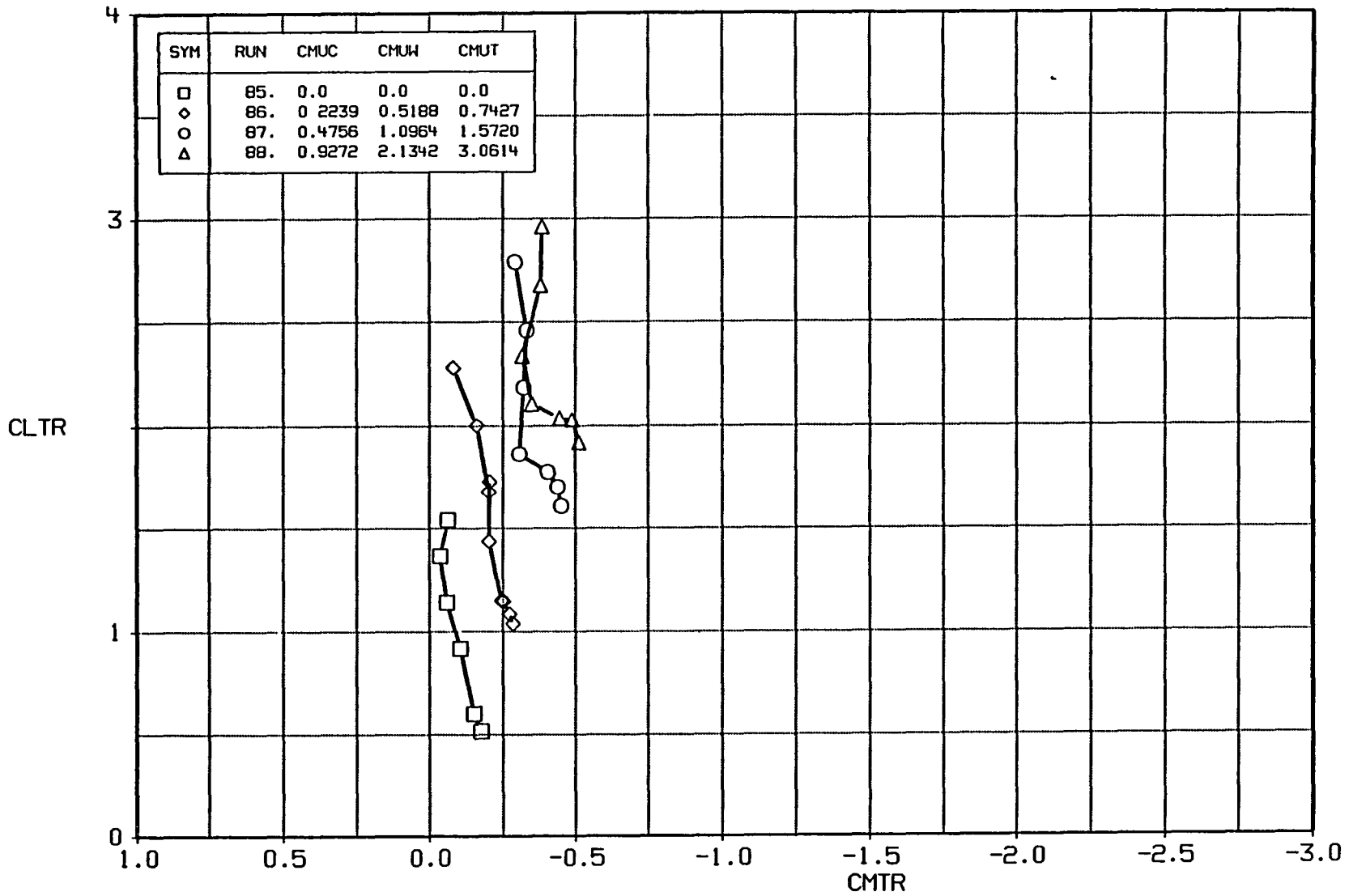


FIGURE A6f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W6V, DELF=45, BN/B=1

A-46

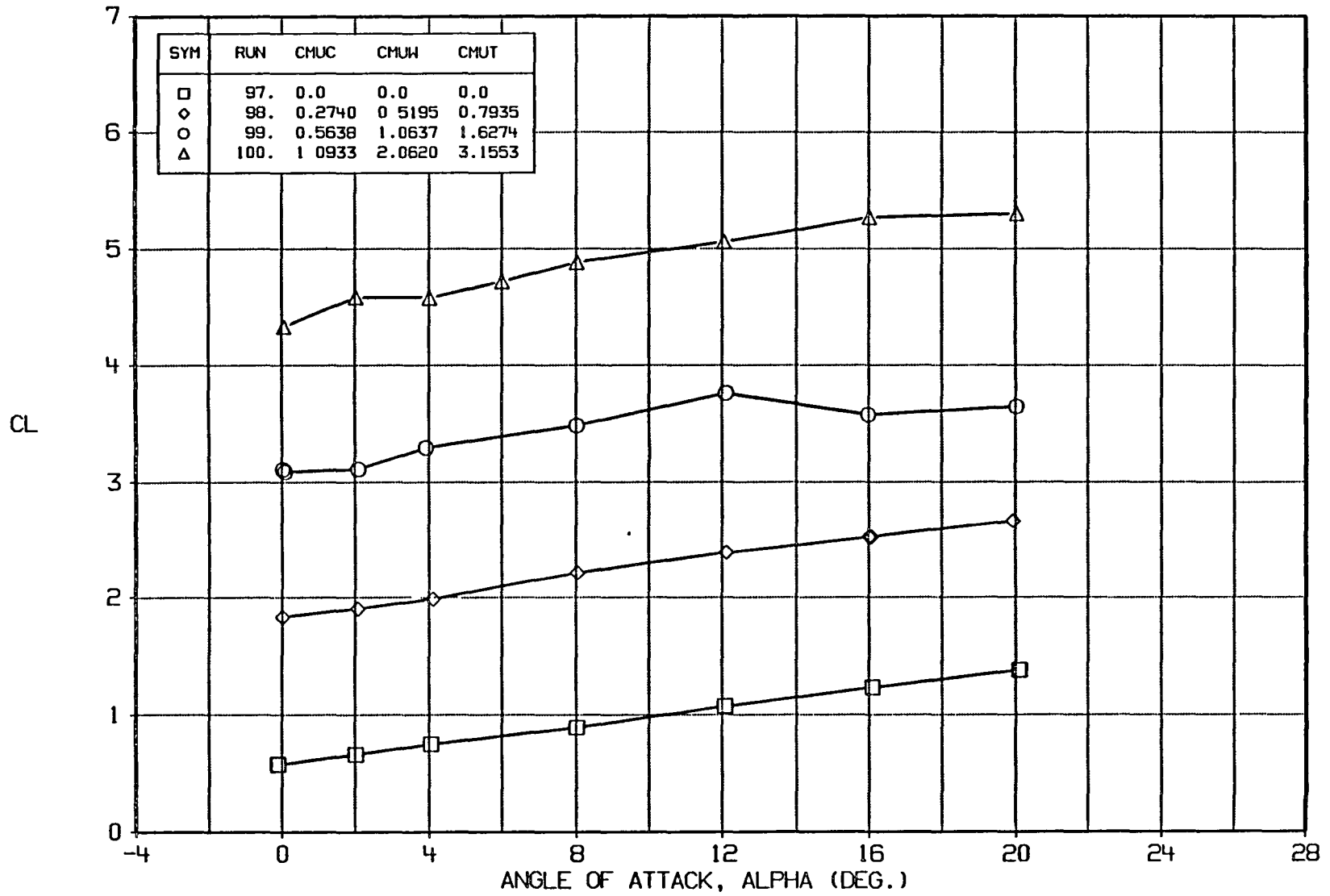


FIGURE A7a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W5V, DELF=45, BN/B=1



A-47

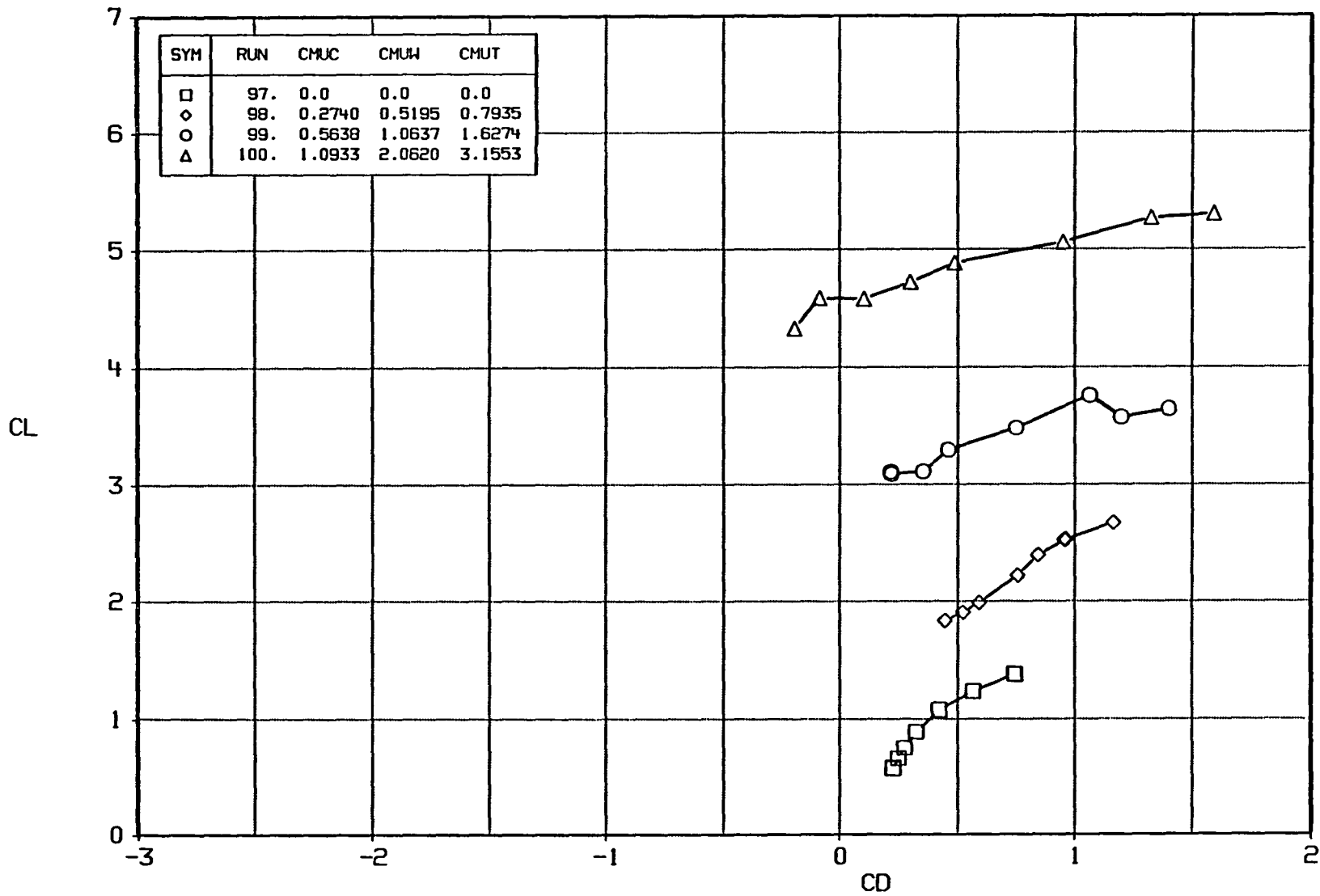


FIGURE A7b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W5V, DELF=45, BN/B=1

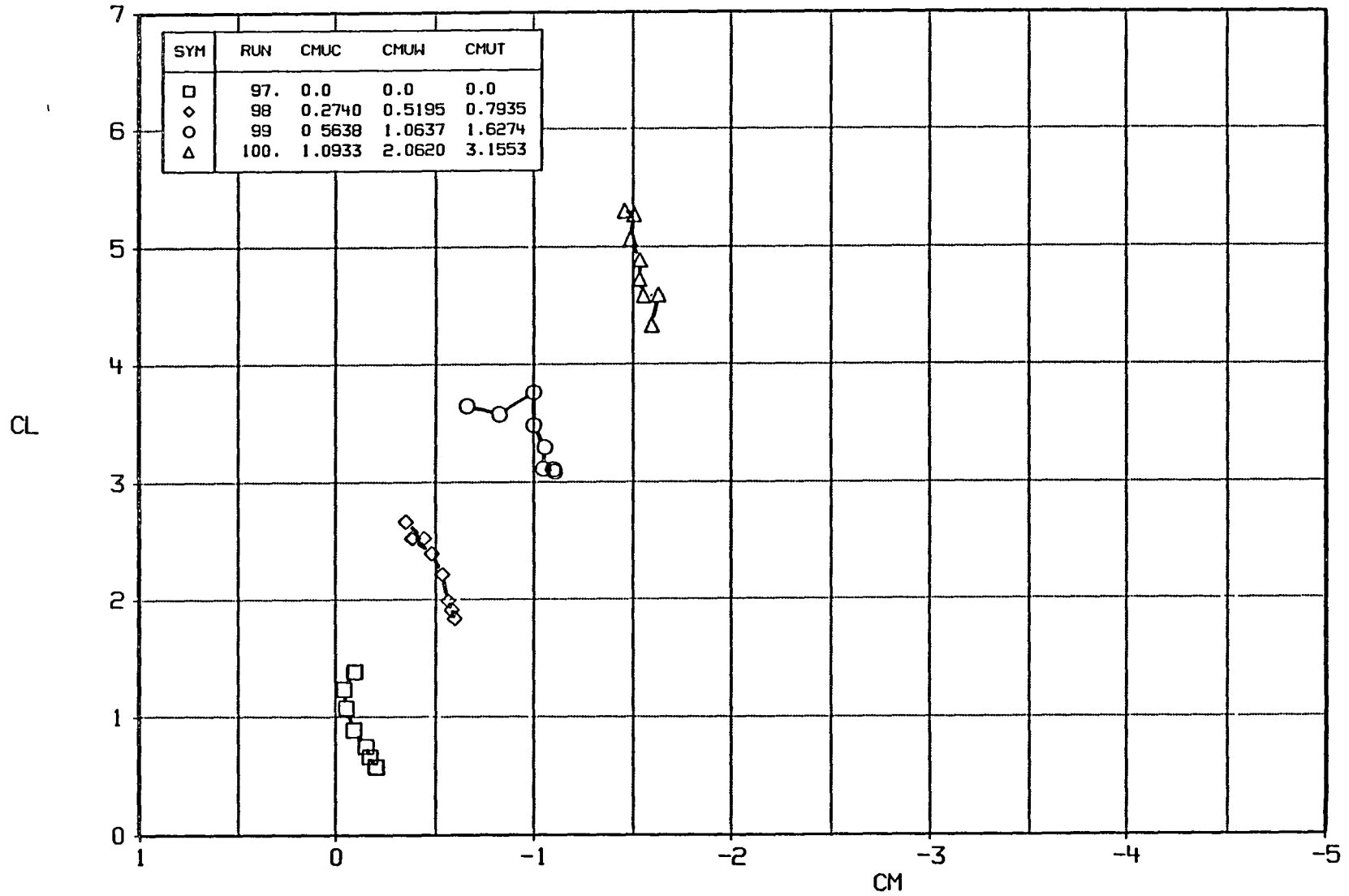


FIGURE A7c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W5V, DELF=45, BN/B=1

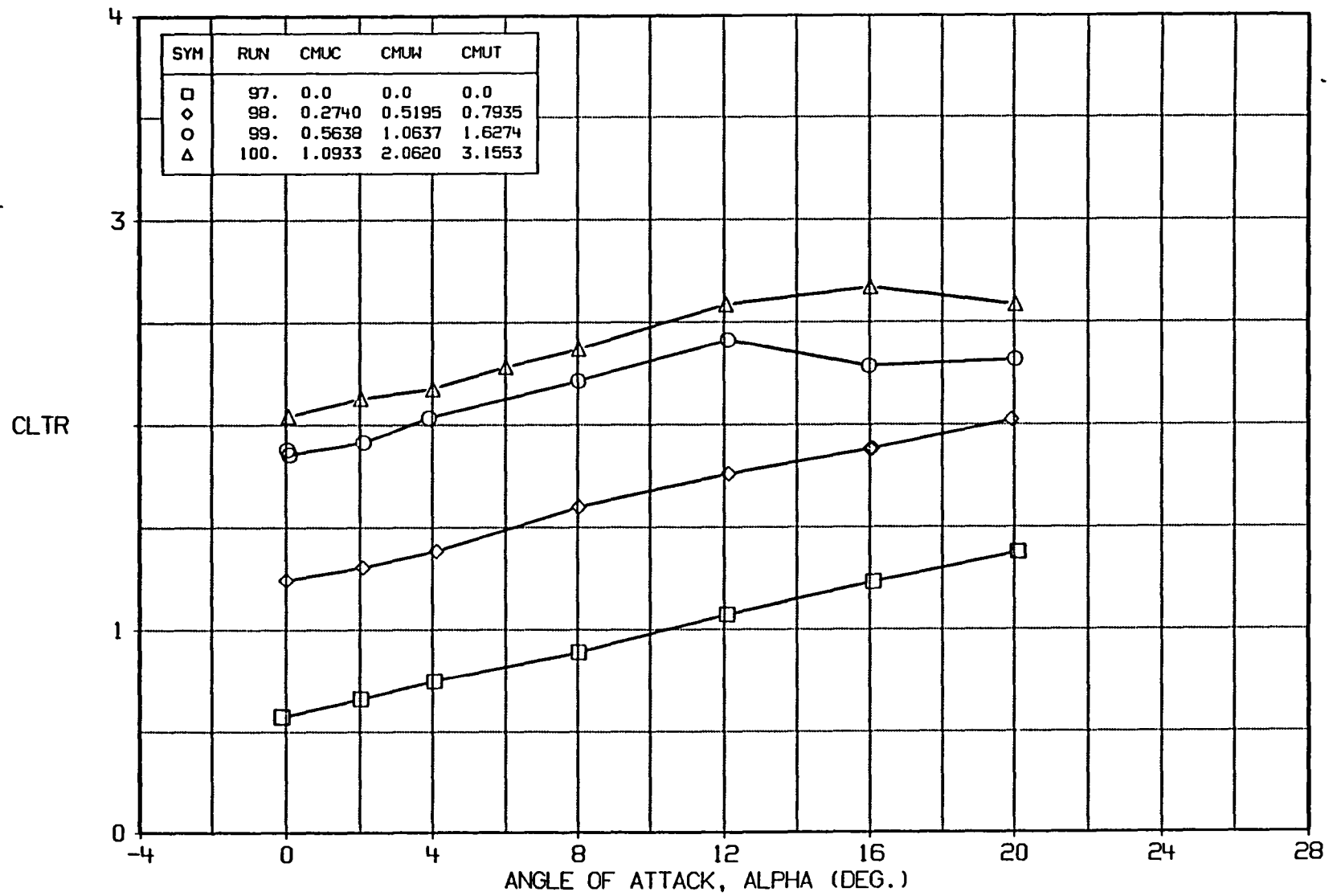


FIGURE A7d BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC2W5V, DELF=45, BN/B=1

A-50

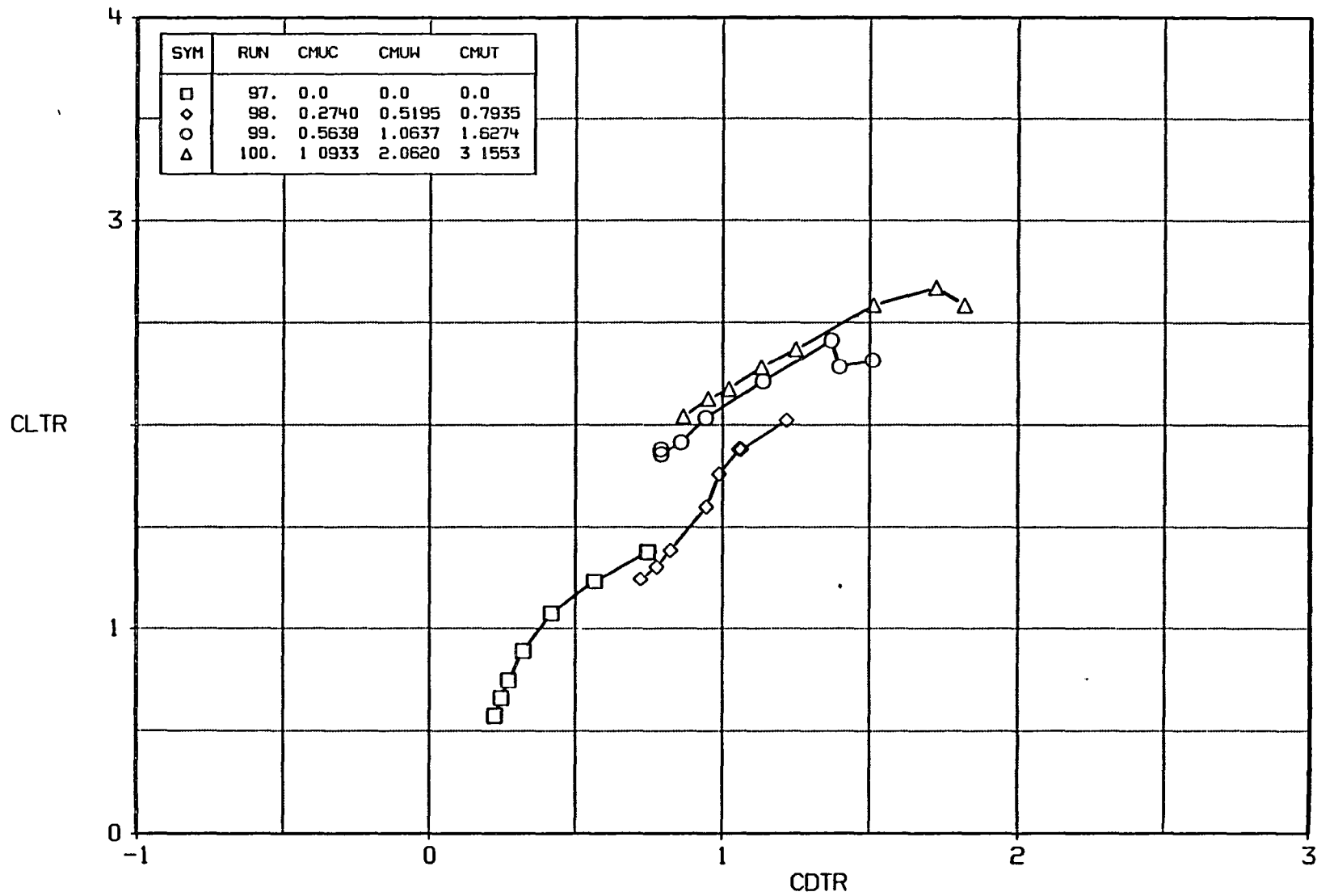


FIGURE A7e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W/5V, DELF=45, BN/B=1

A-51

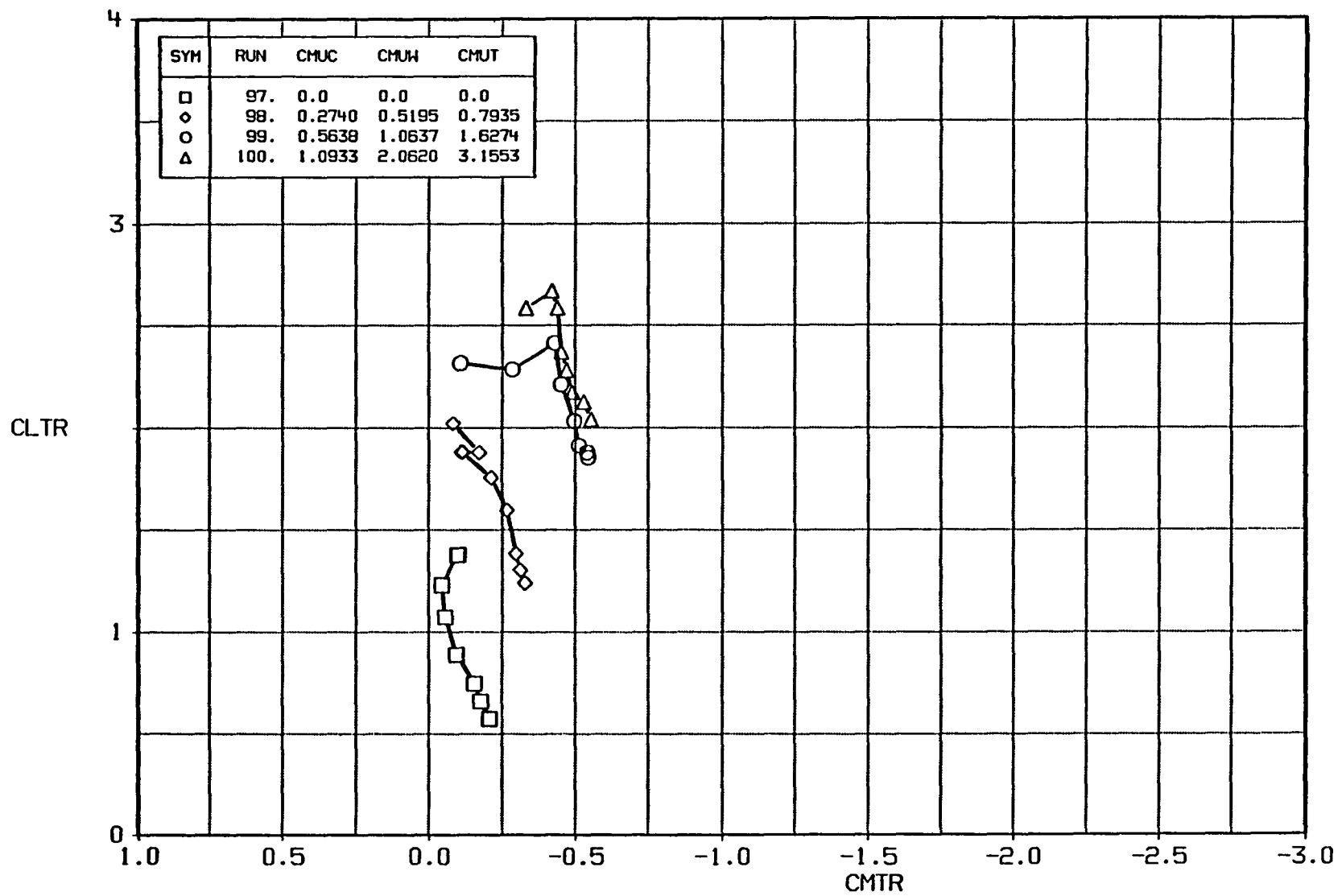


FIGURE A7f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W5V, DELF=45, BN/B=1

A-52

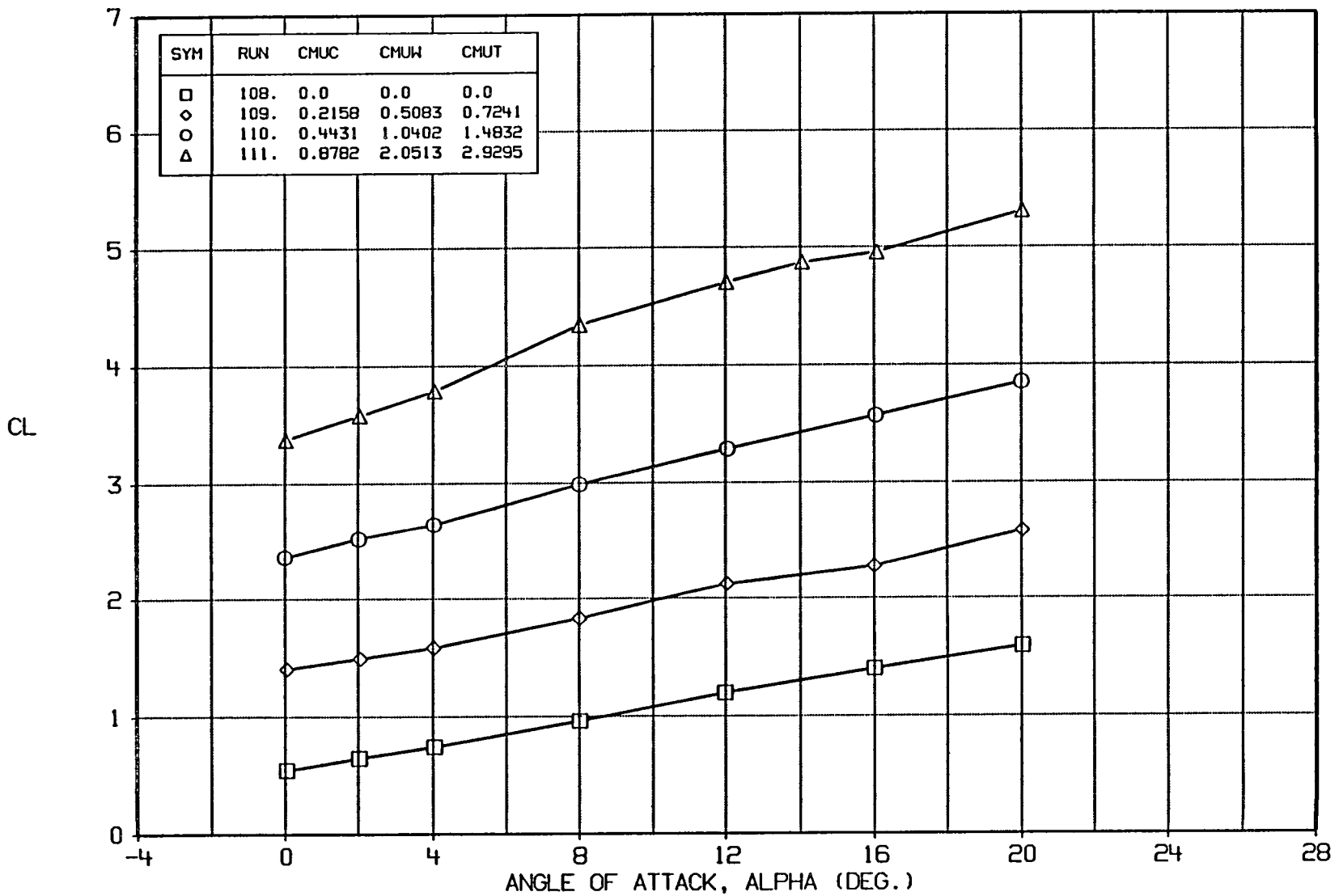


FIGURE A8a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC3W6V, DELF=45, BN/B=1

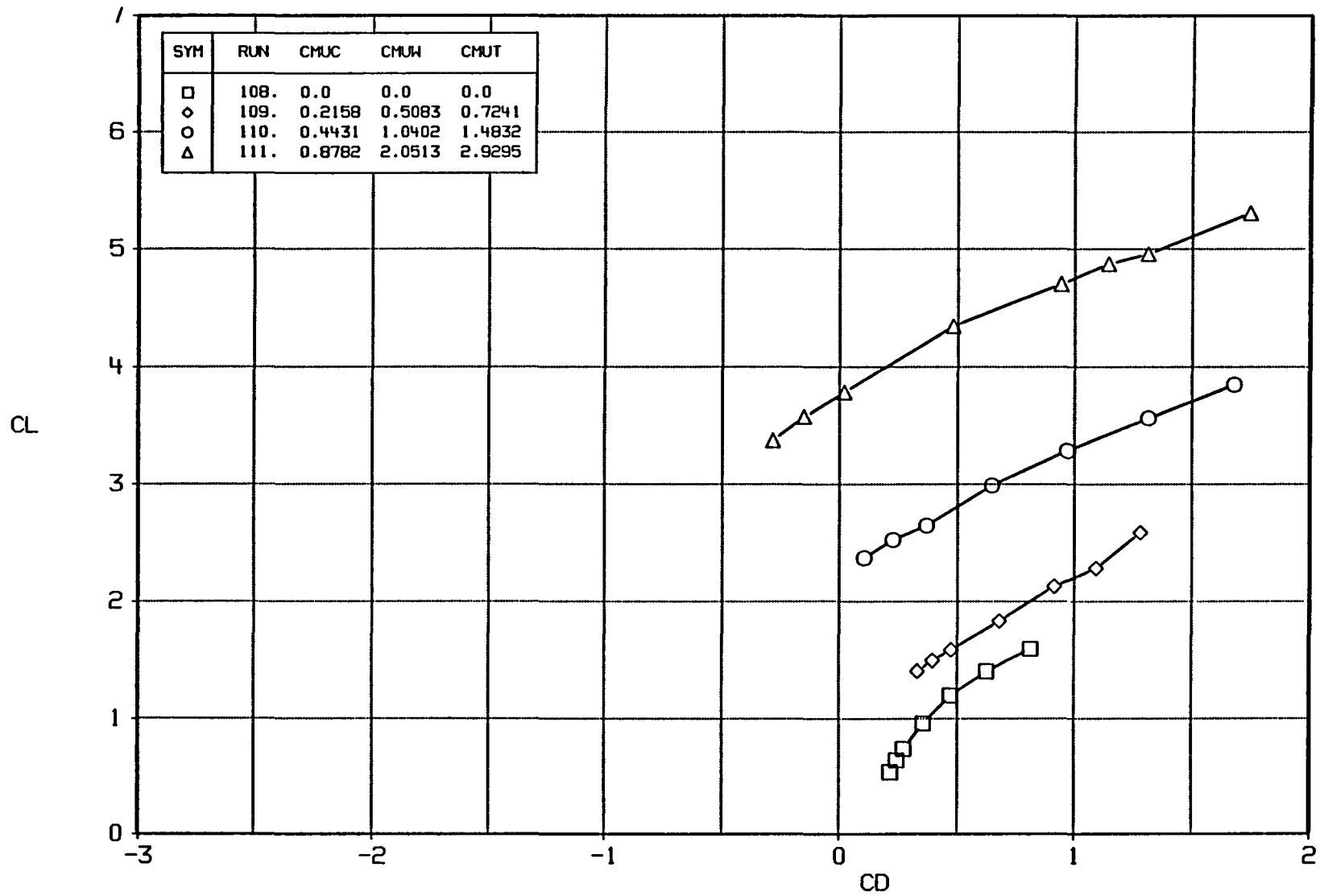


FIGURE A8b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC3W6V, DELF=45, BN/B=1

A-54

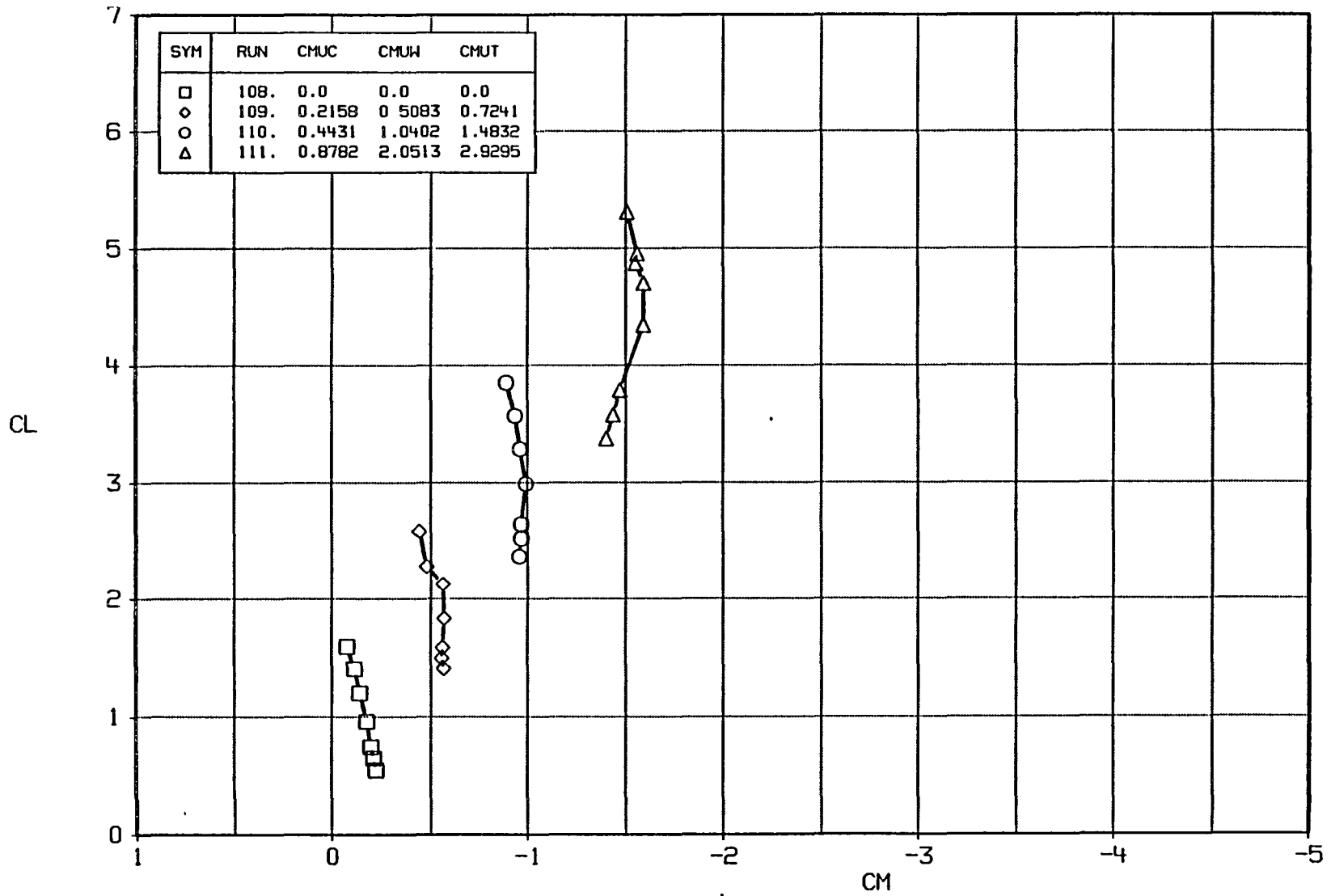


FIGURE A8c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC3W6V, DELF=45, BN/B=1



A-55

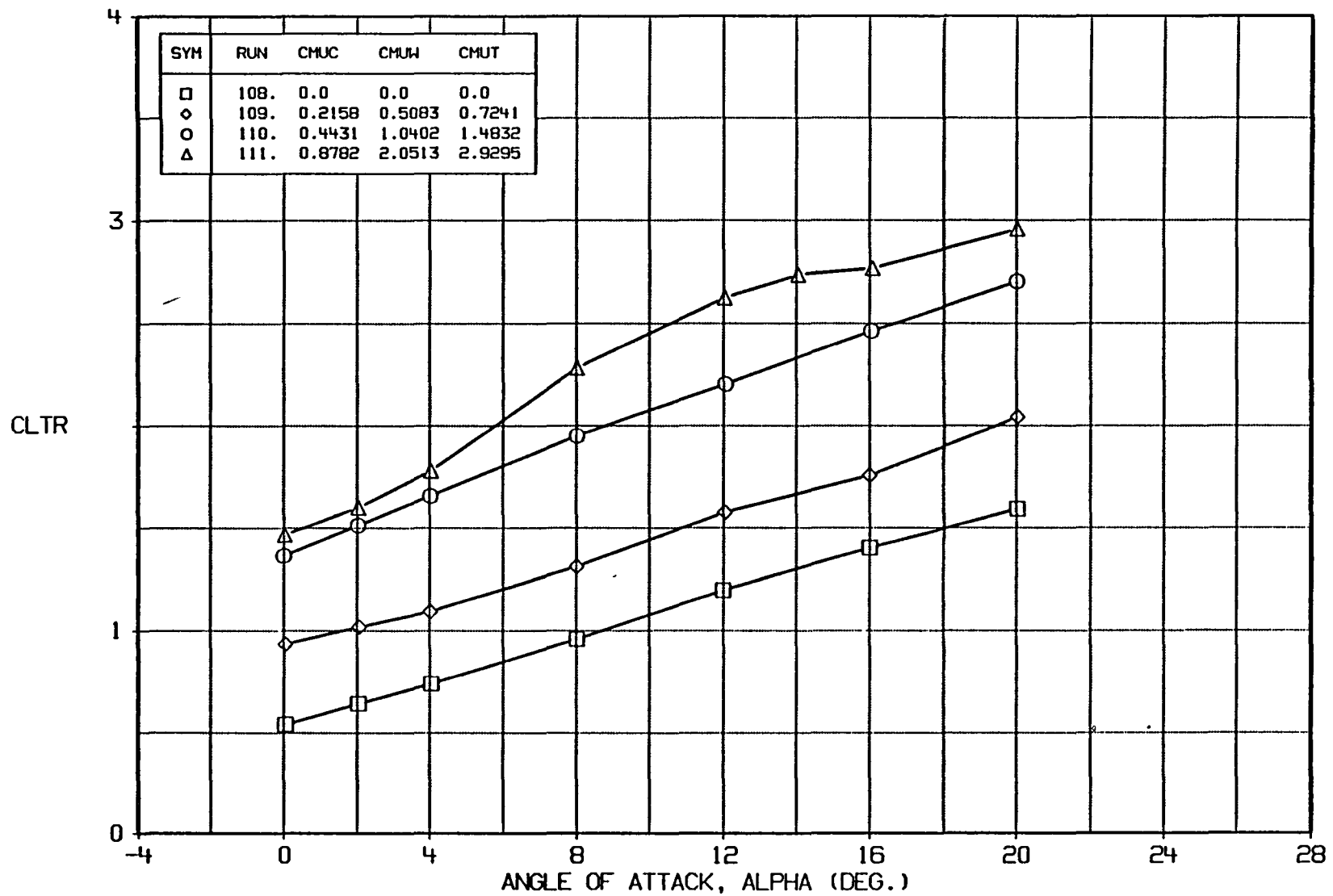


FIGURE A8d BASIC DATA EFFECT OF CMU  
CONFIGURATION BC3W6V, DELF=45, BN/B=1

A-56

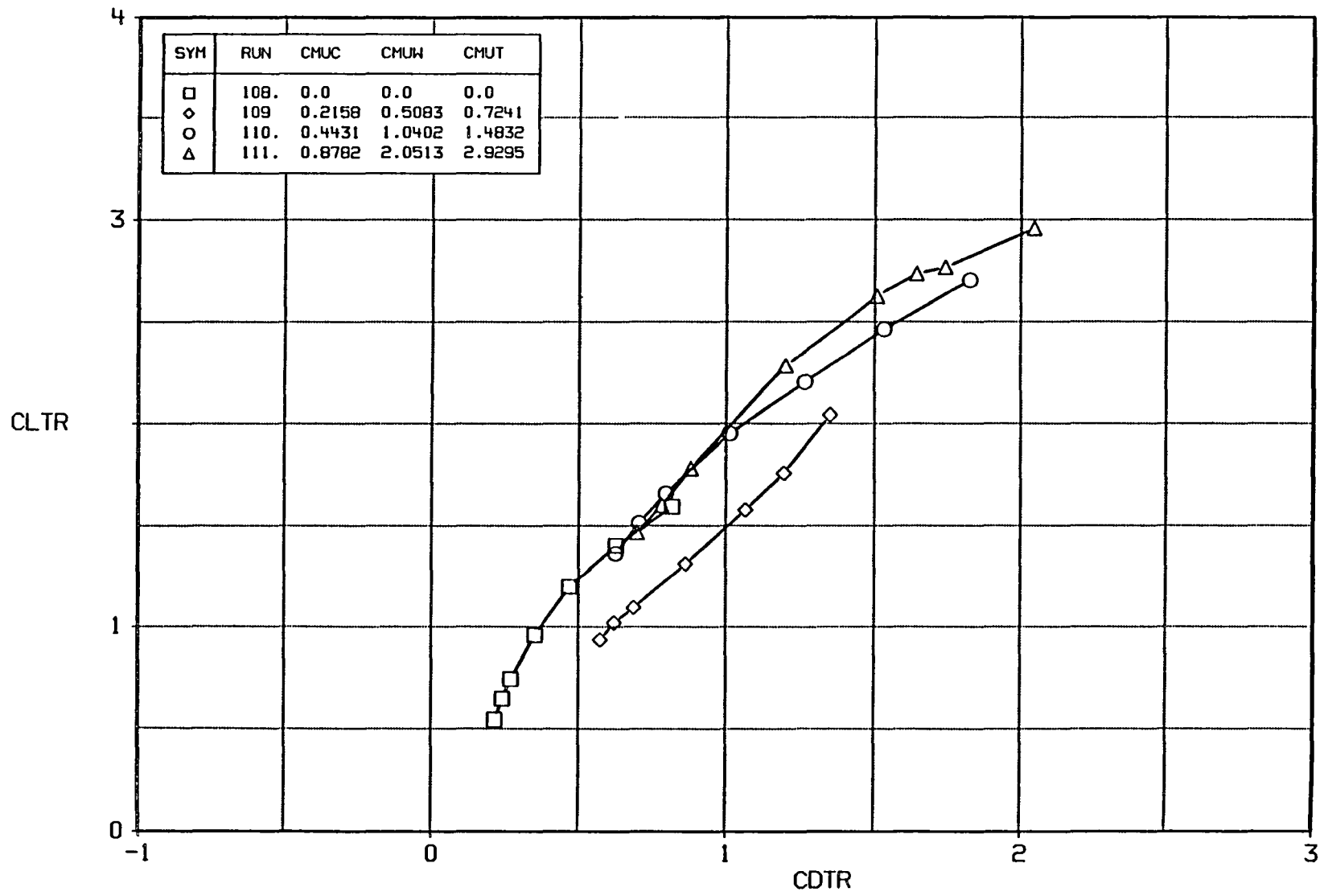


FIGURE A8e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC3W6V, DELF=45, BN/B=1

A-57

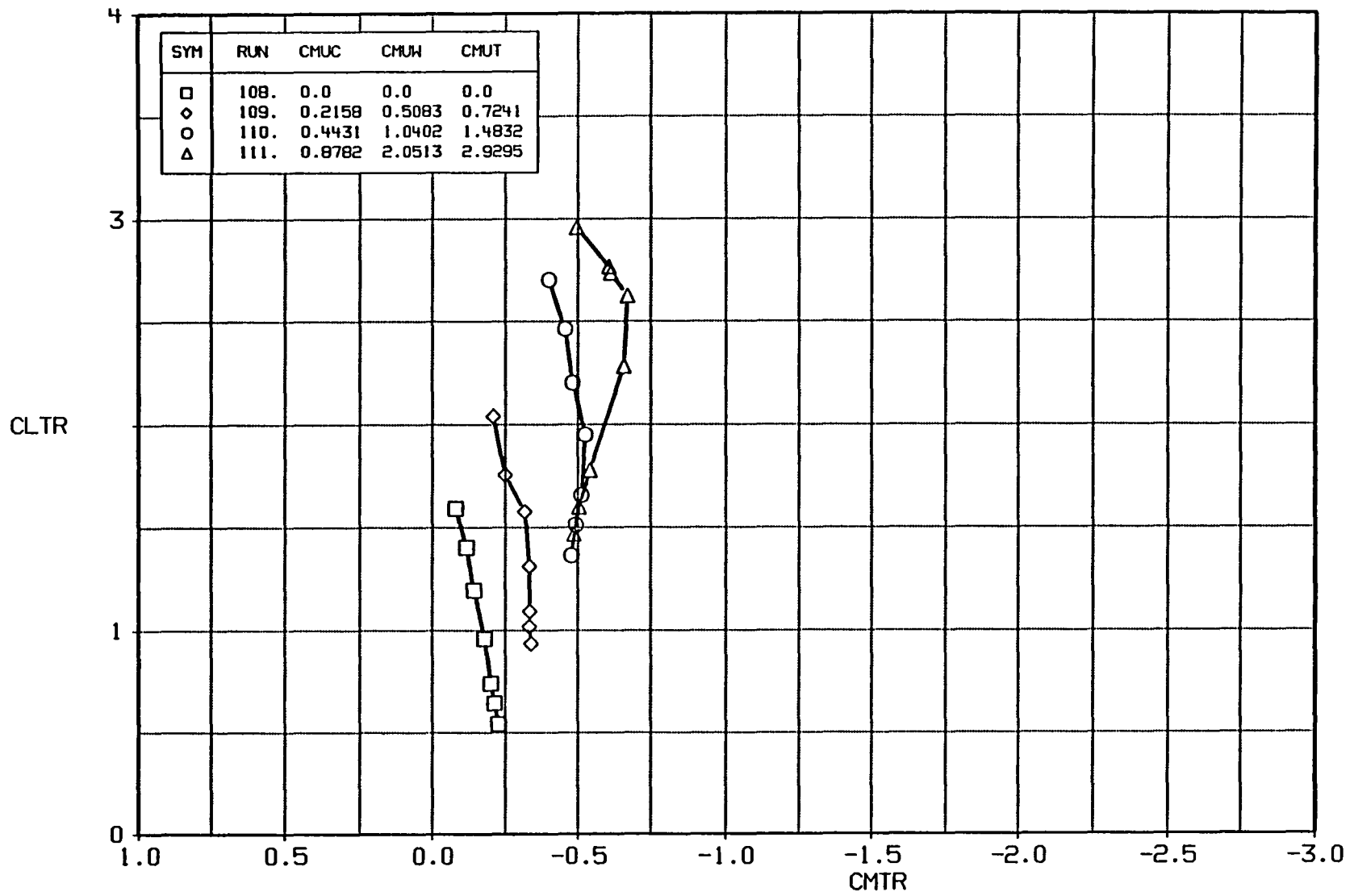


FIGURE A8f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC3W6V, DELF=45, BN/B=1

A-58

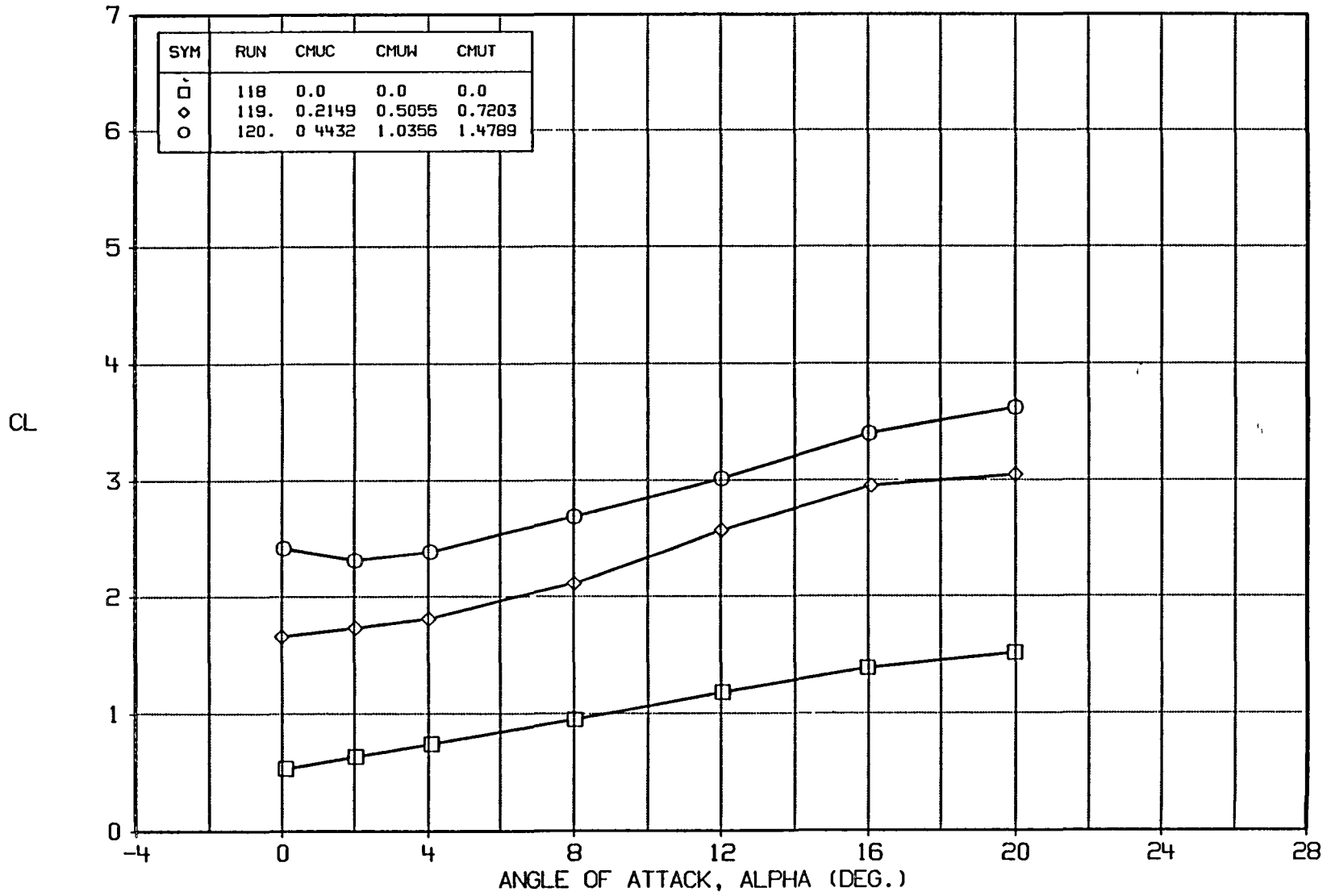


FIGURE A9a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10

A-59

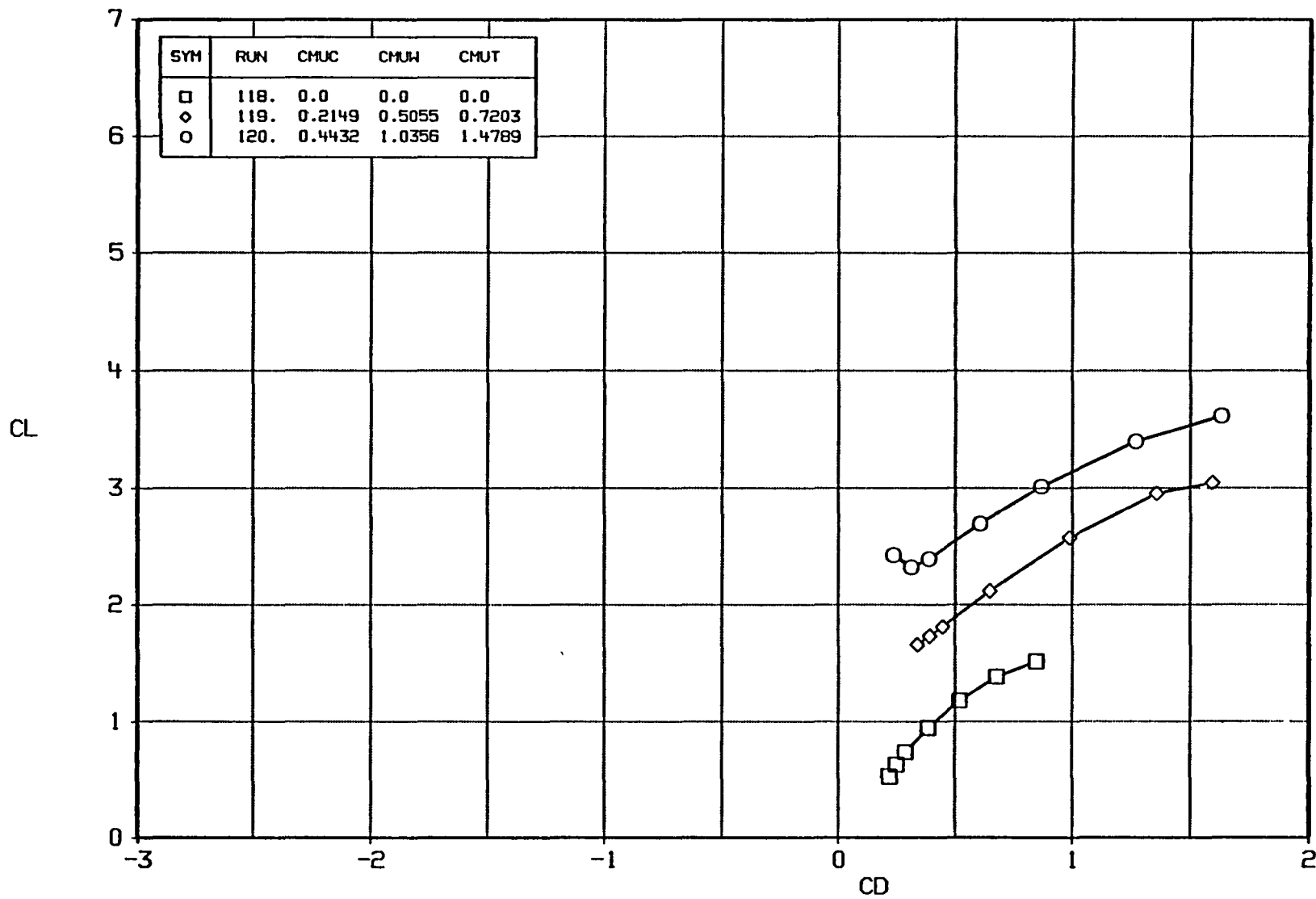


FIGURE A9b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10

A-60

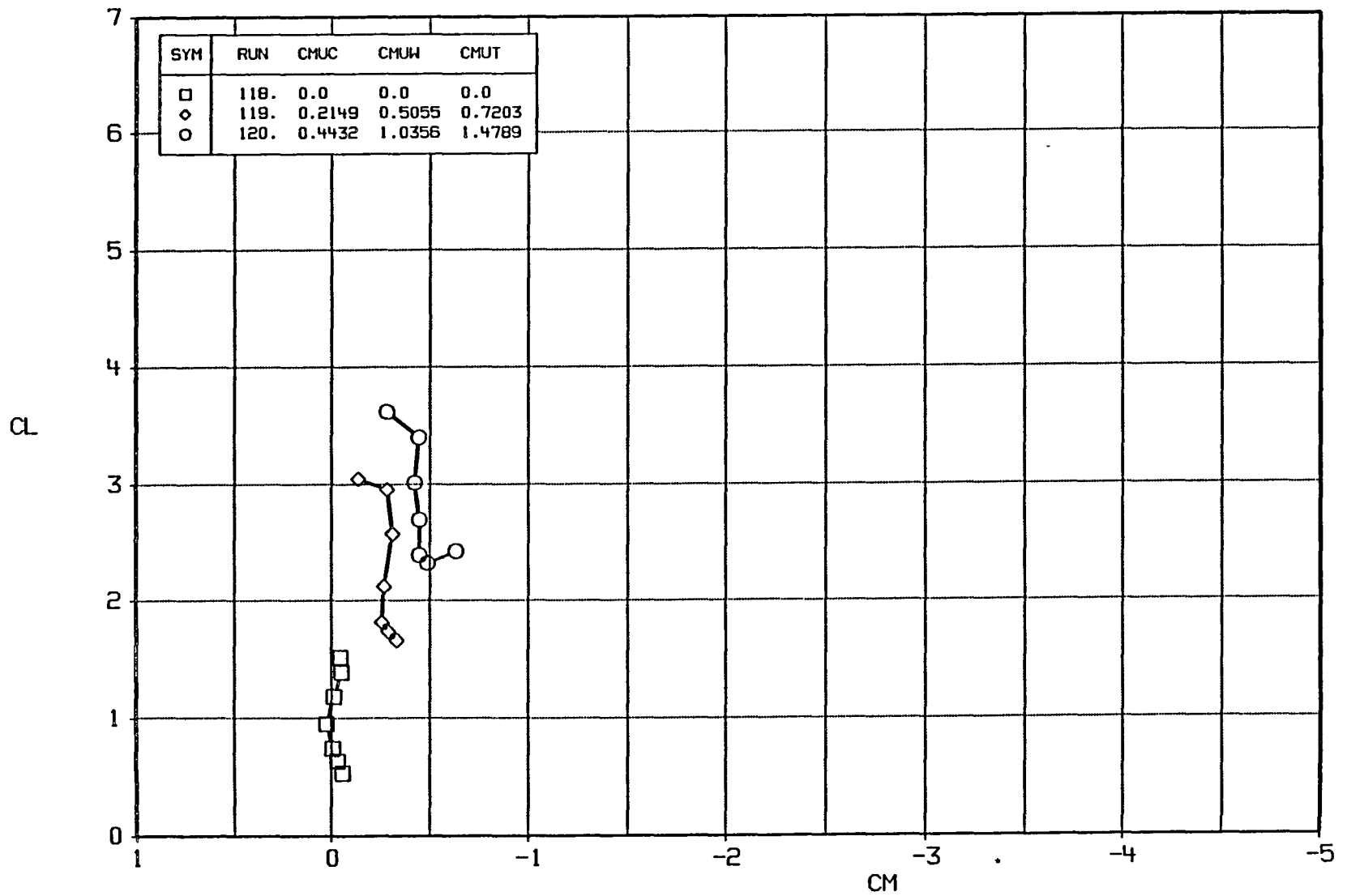


FIGURE A9c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10

A-61

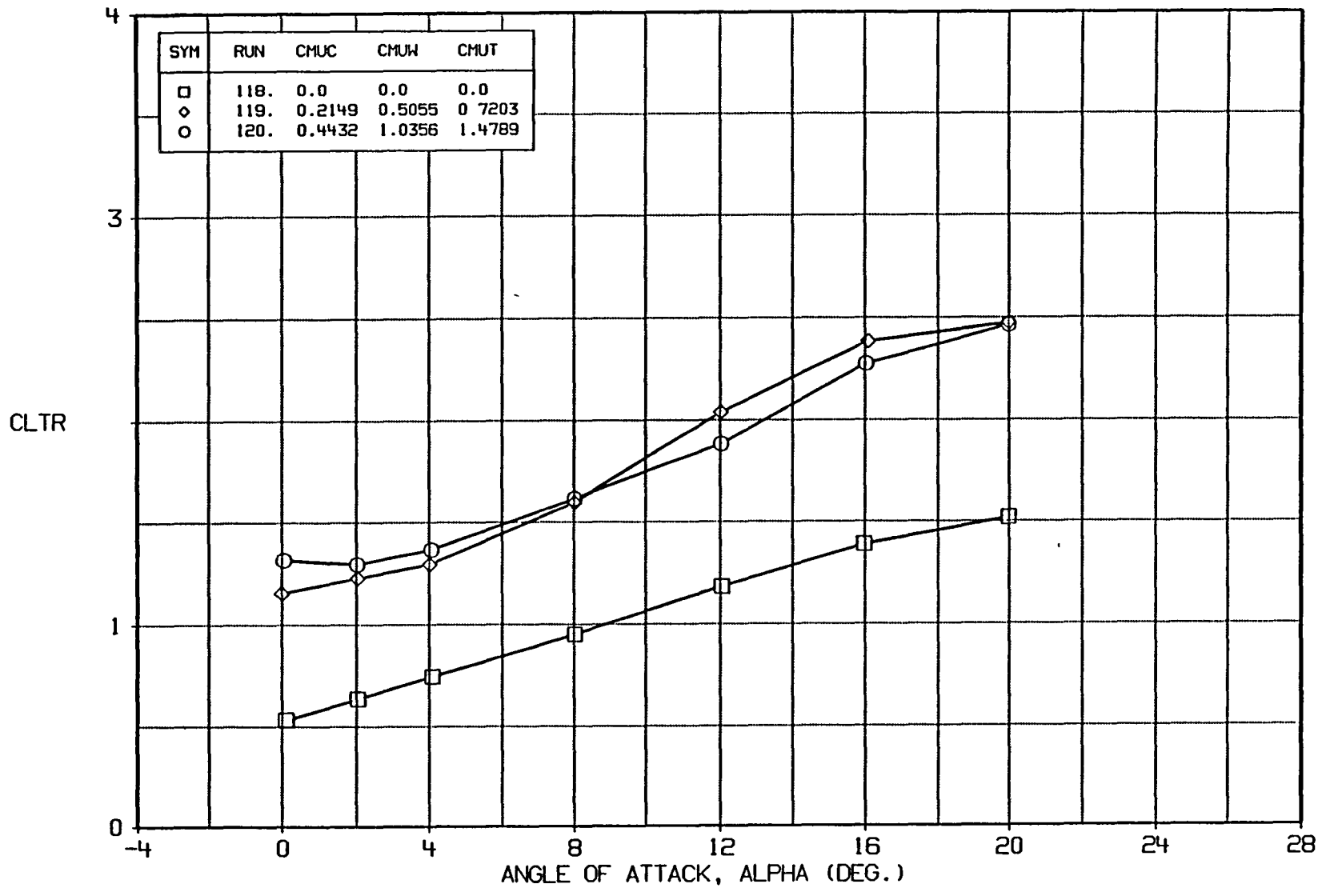


FIGURE A9d BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10

A-62

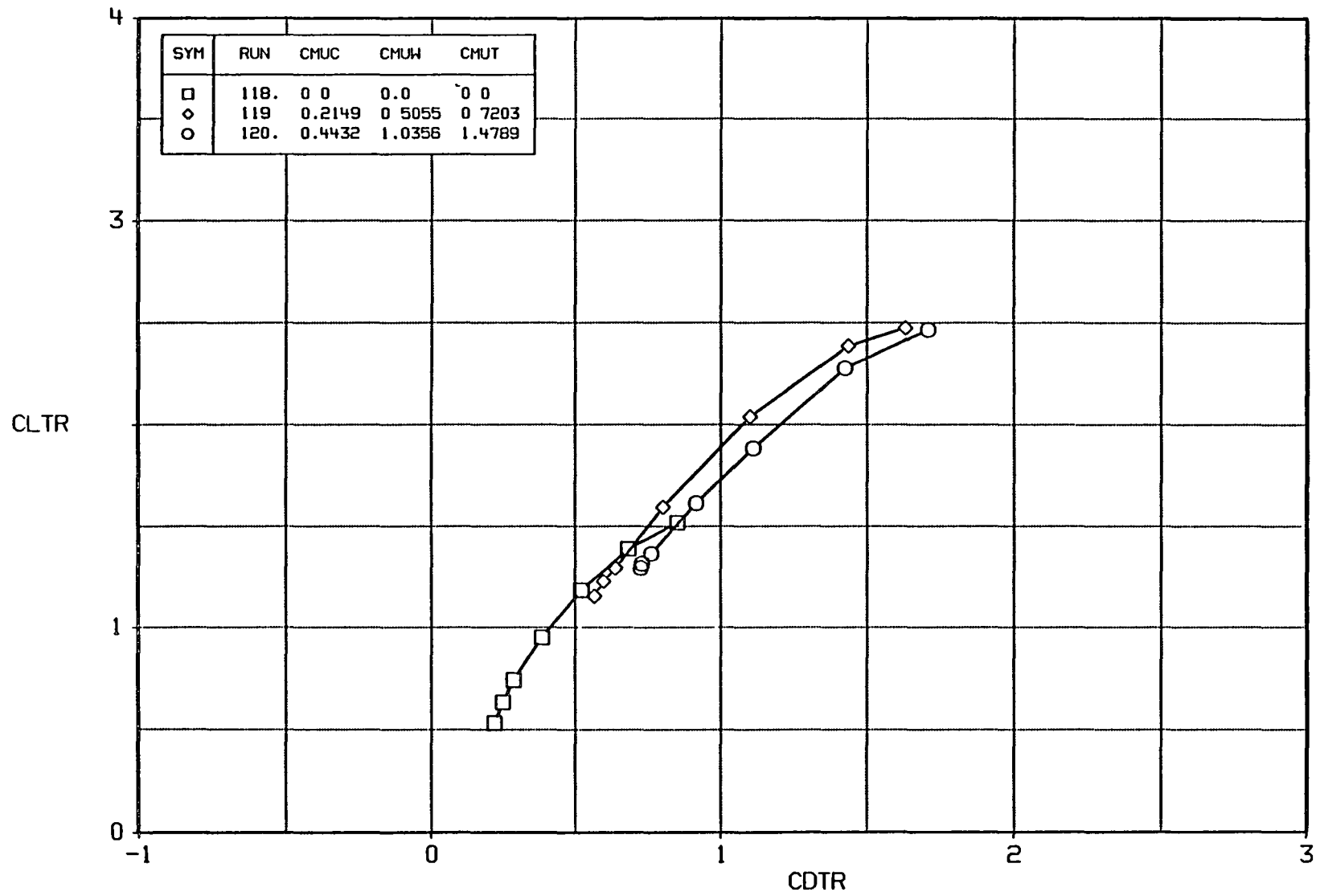


FIGURE A9e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10



A-63

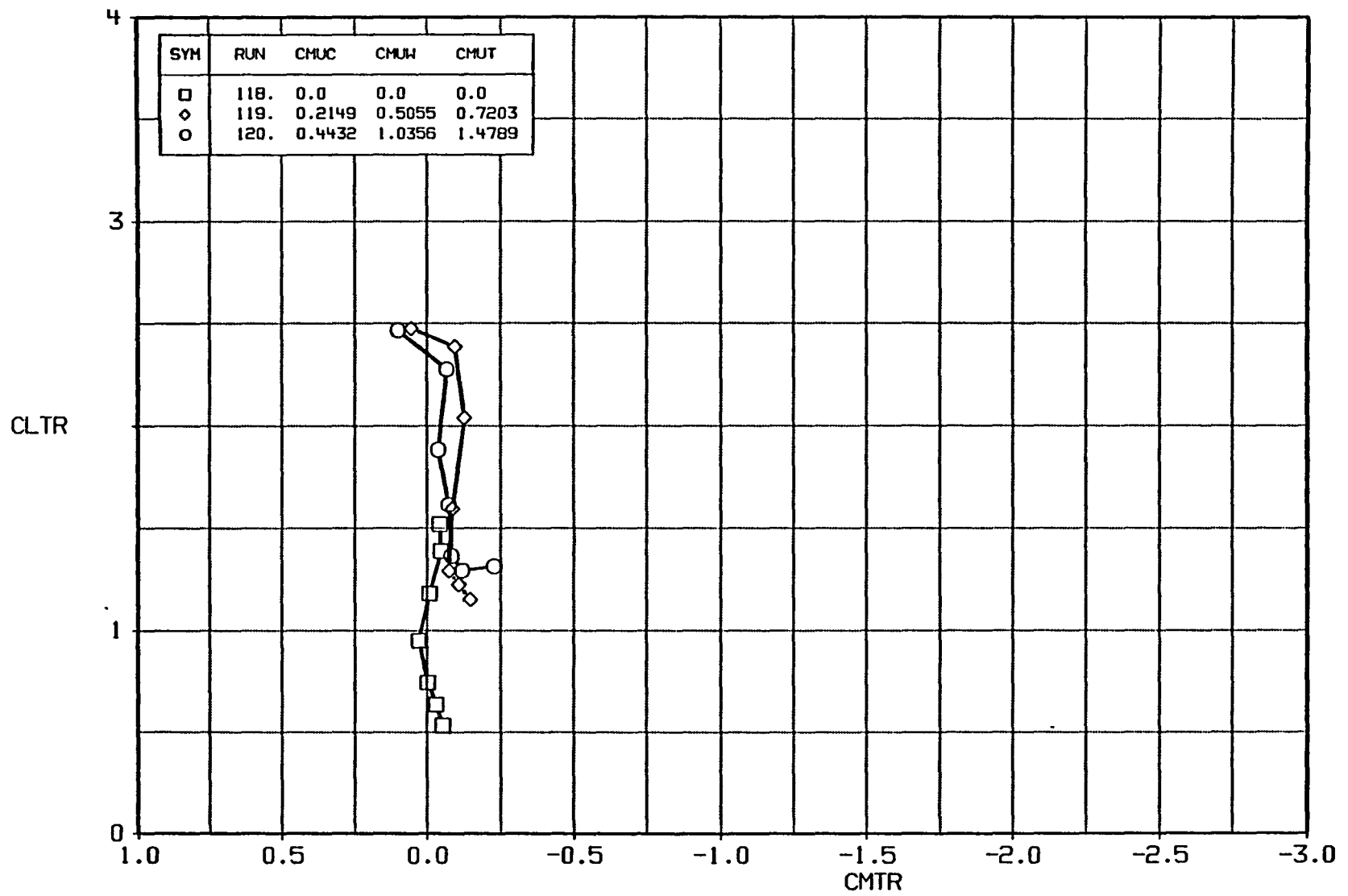


FIGURE A9f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10

A-64

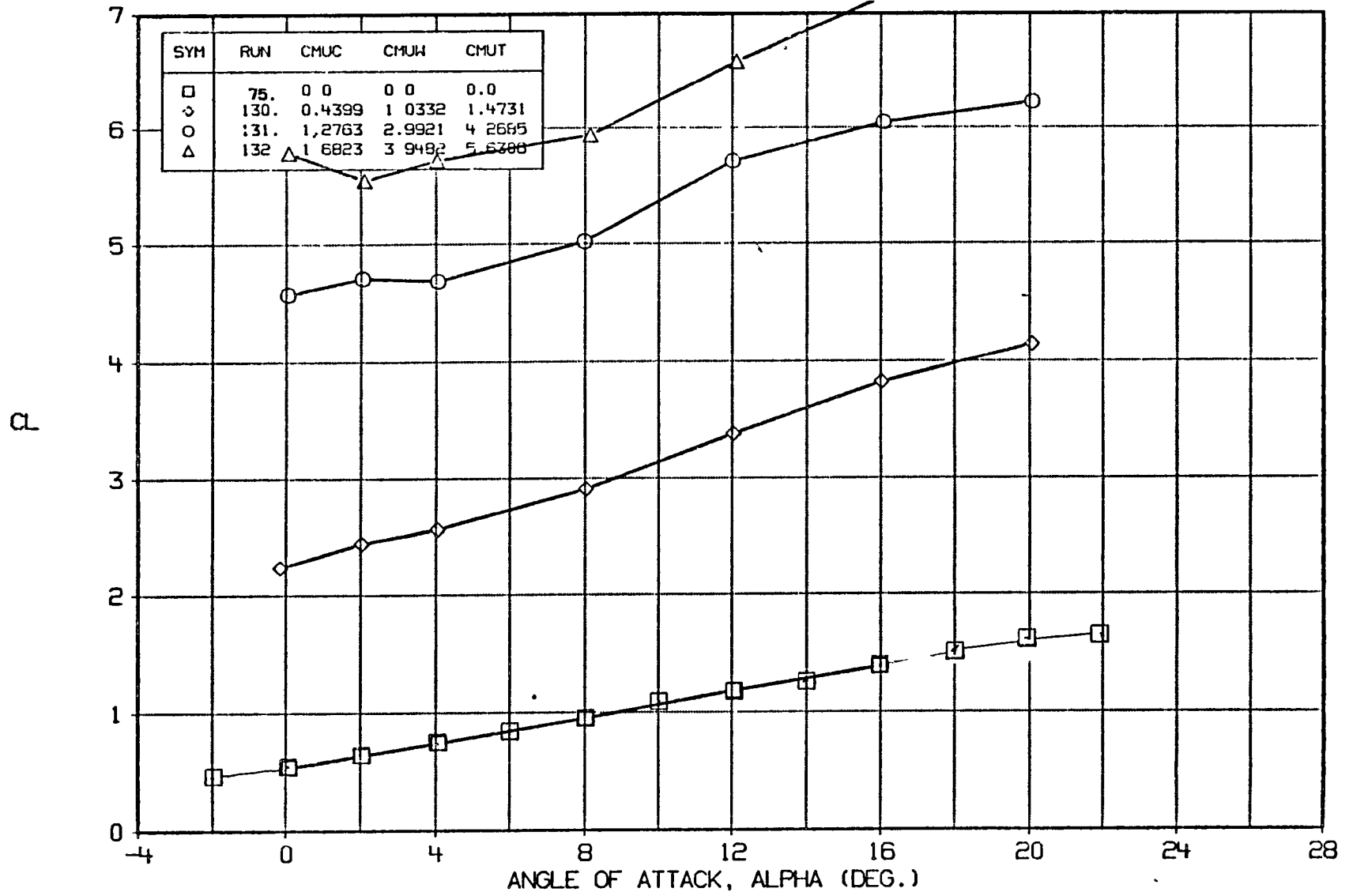


FIGURE A10a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-65

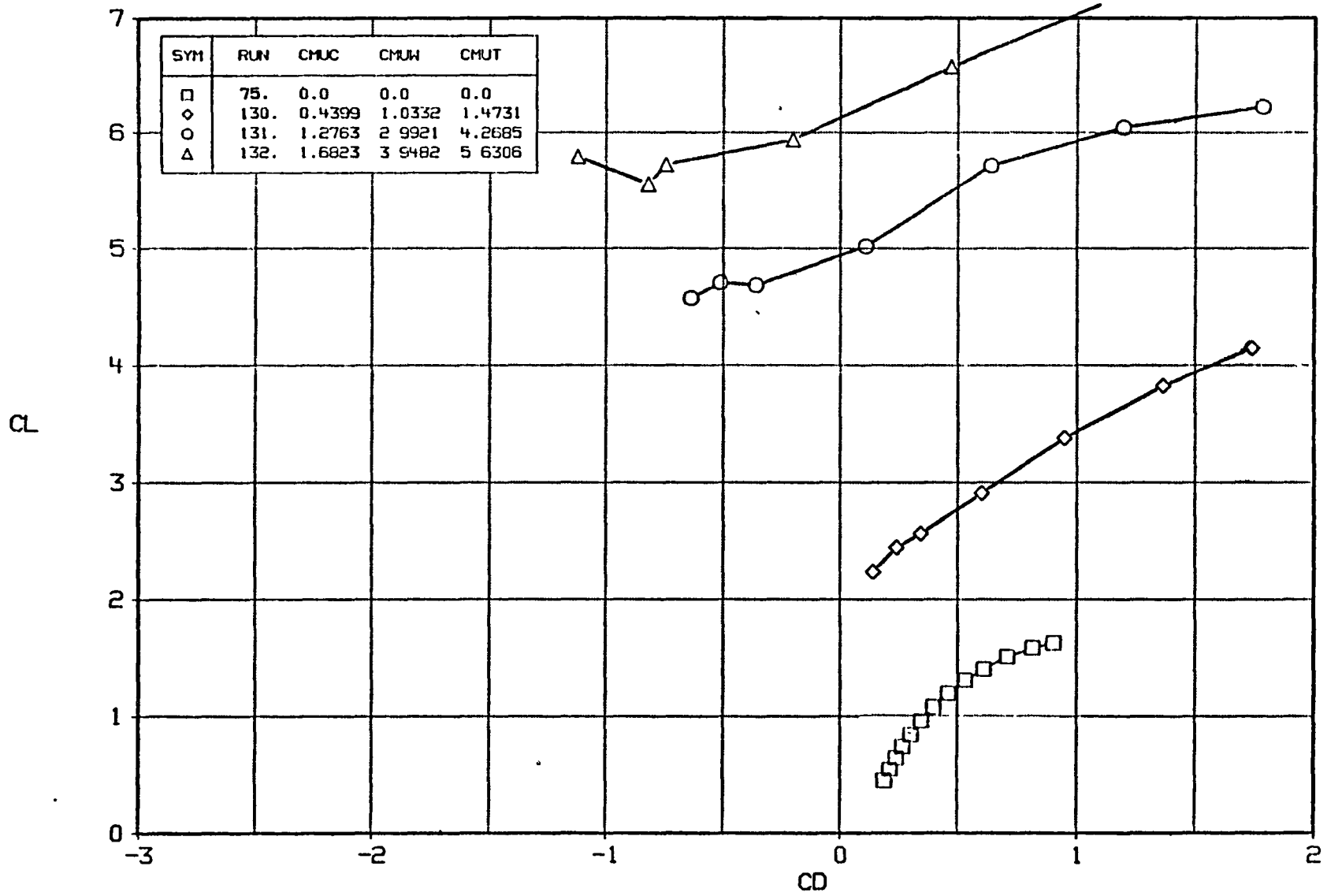


FIGURE A10b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-66

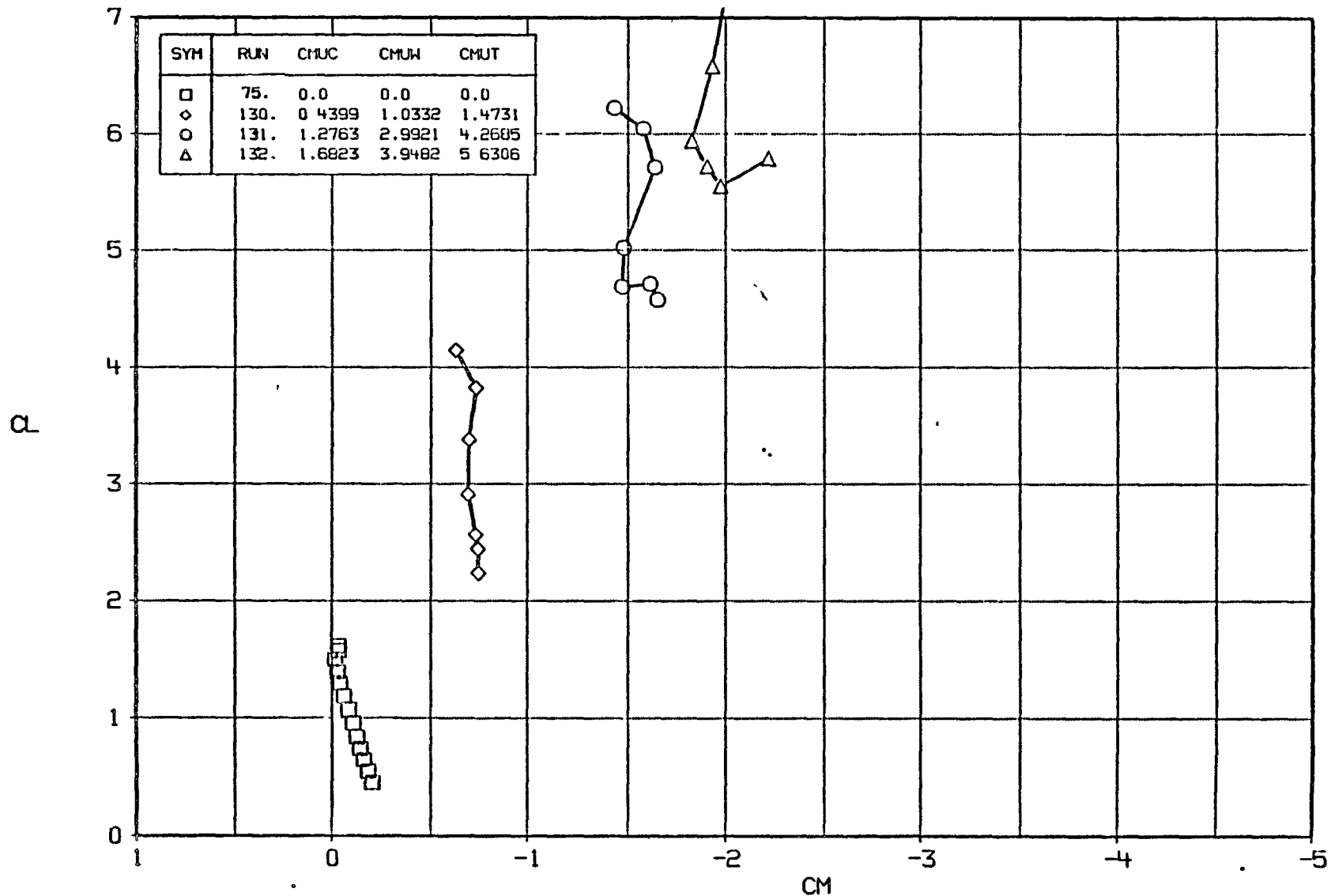


FIGURE A10c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-67

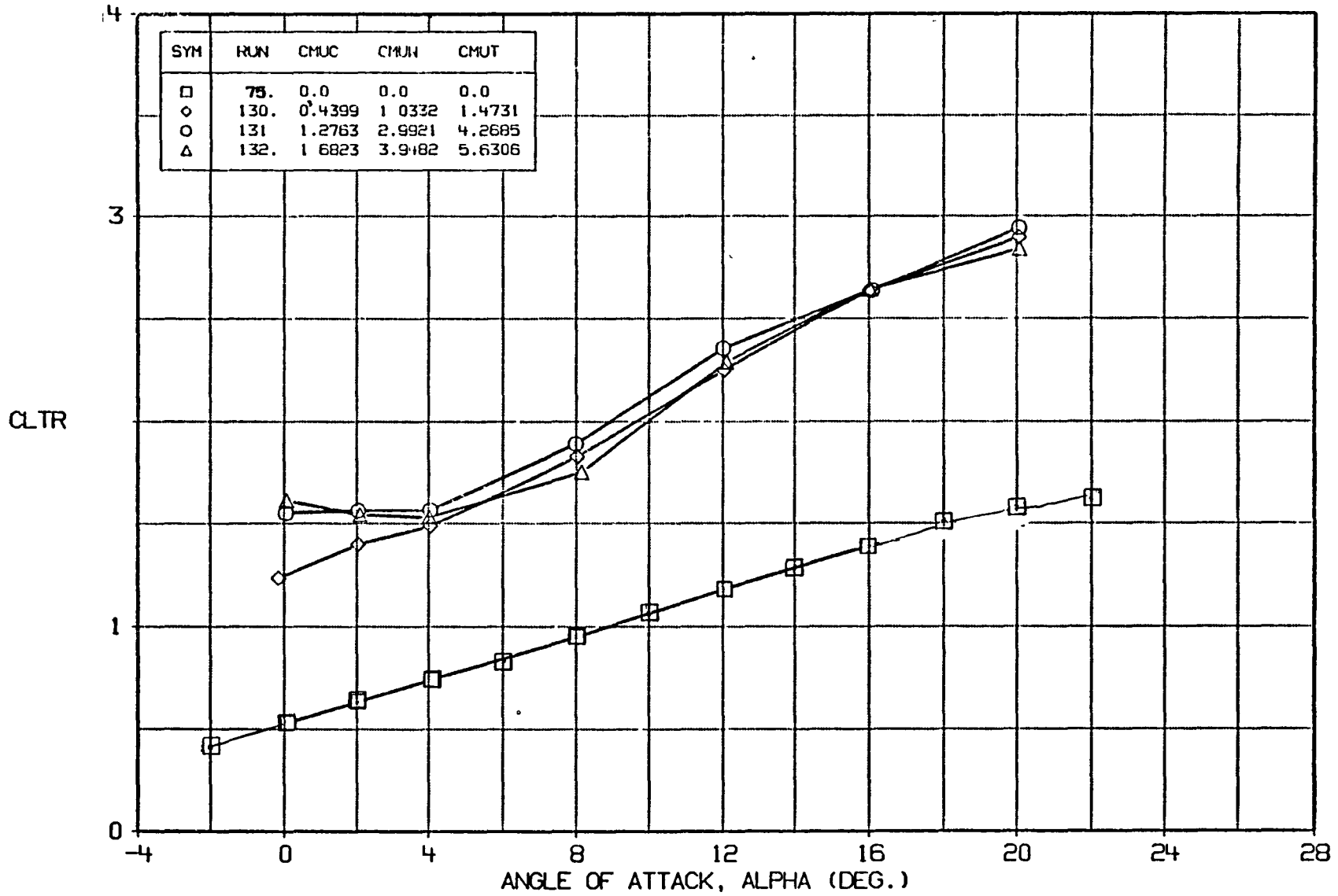


FIGURE A10d BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-68

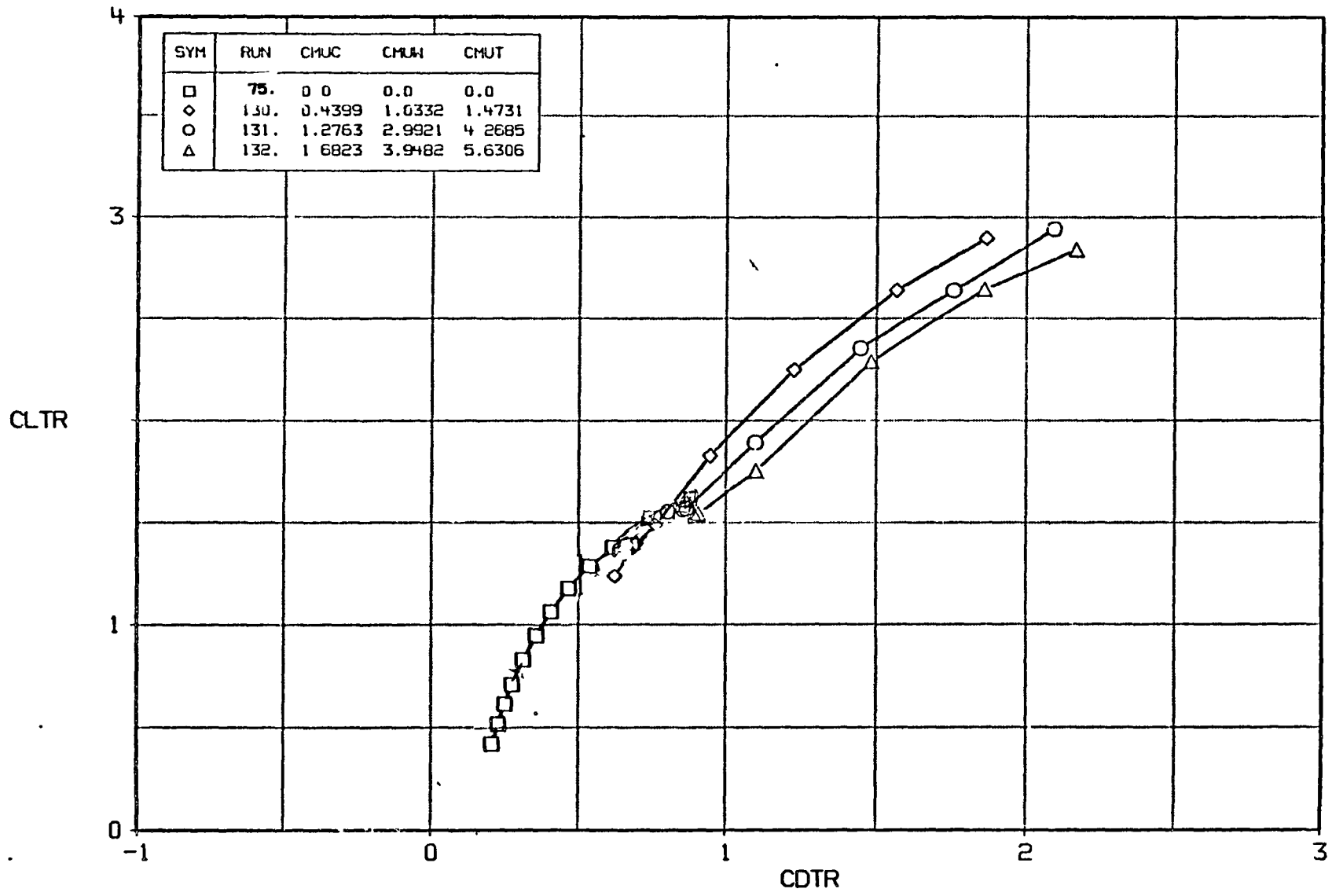


FIGURE A10e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

69-V

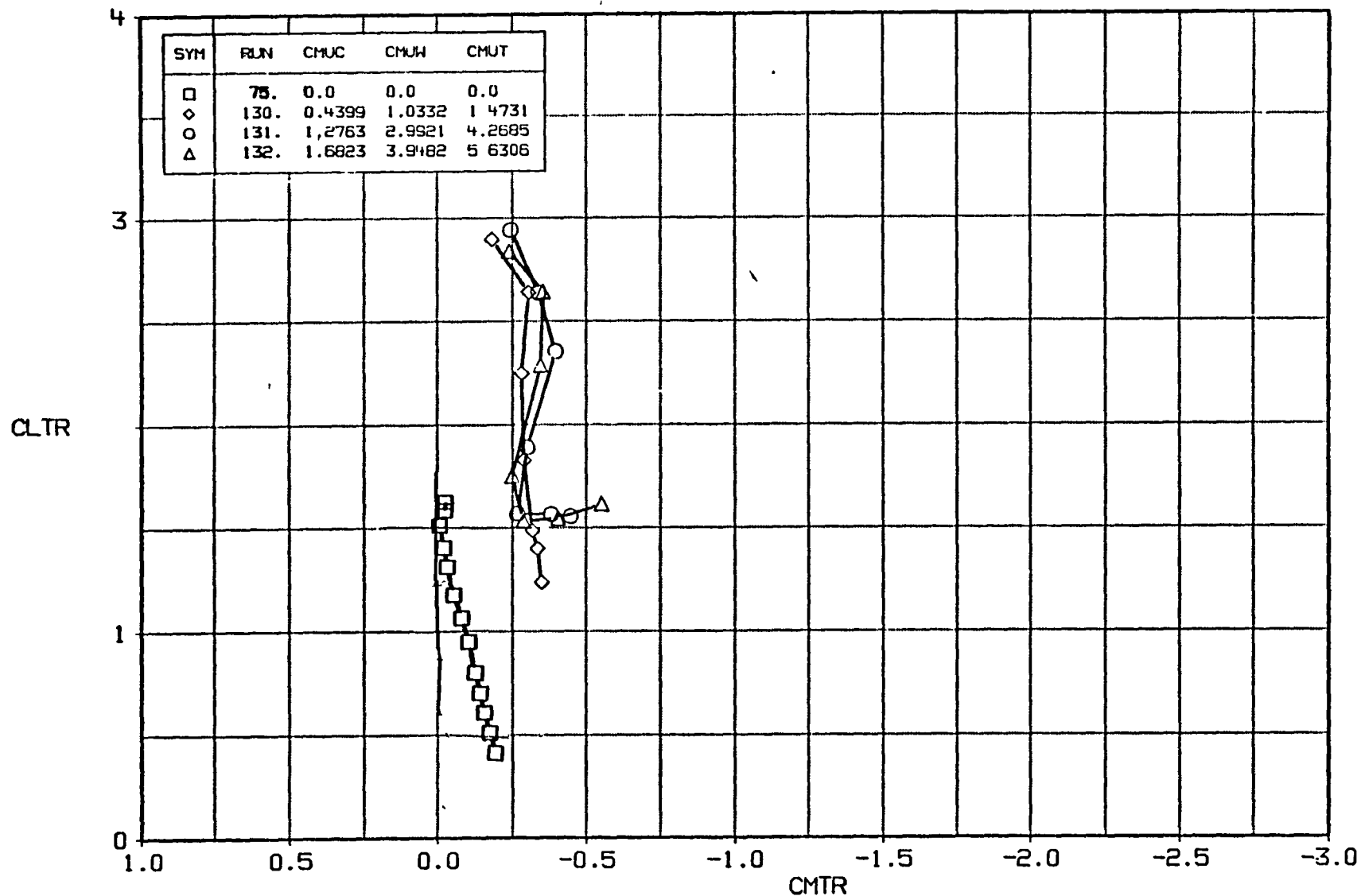


FIGURE A10f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-70

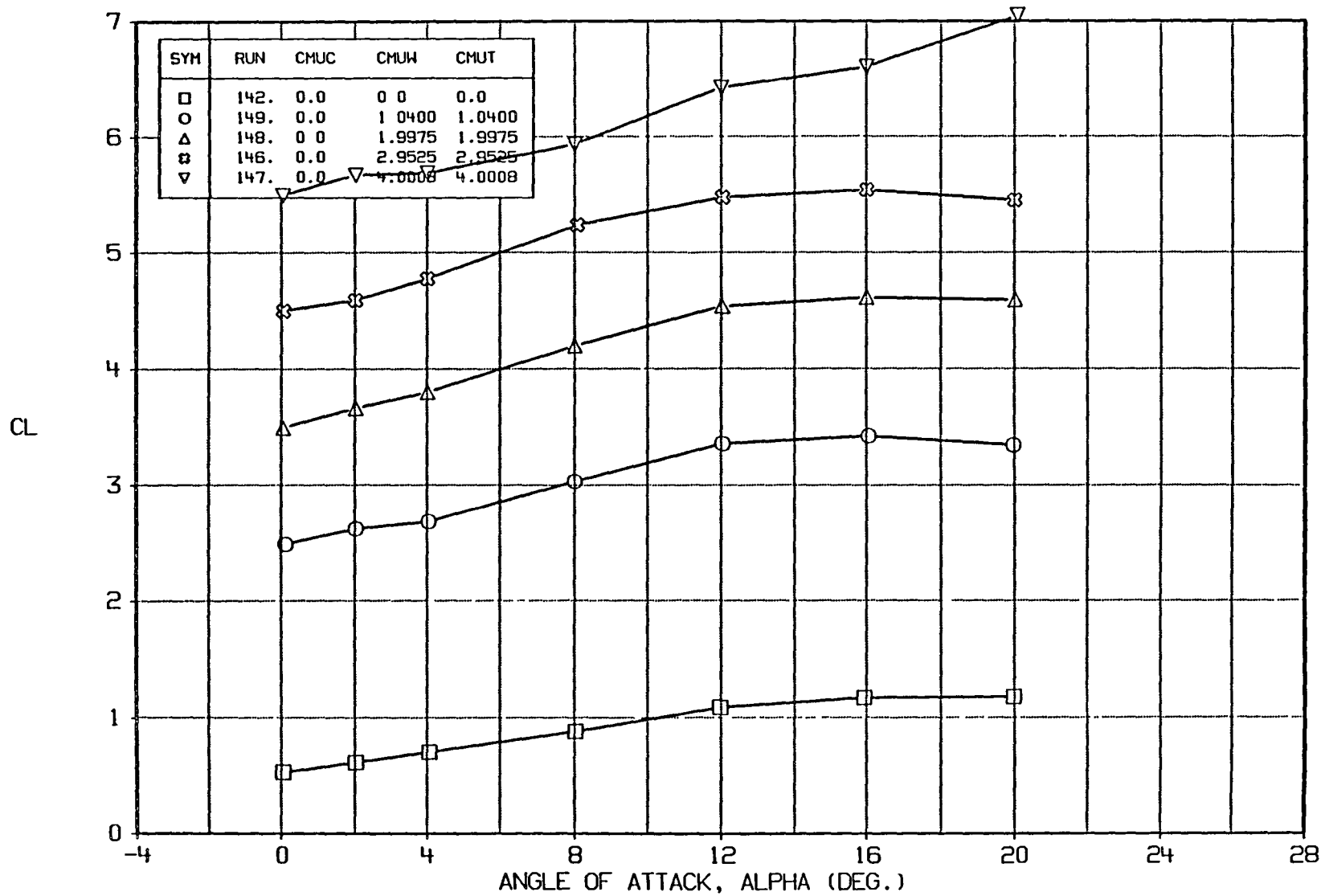


FIGURE A11a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1



A-71

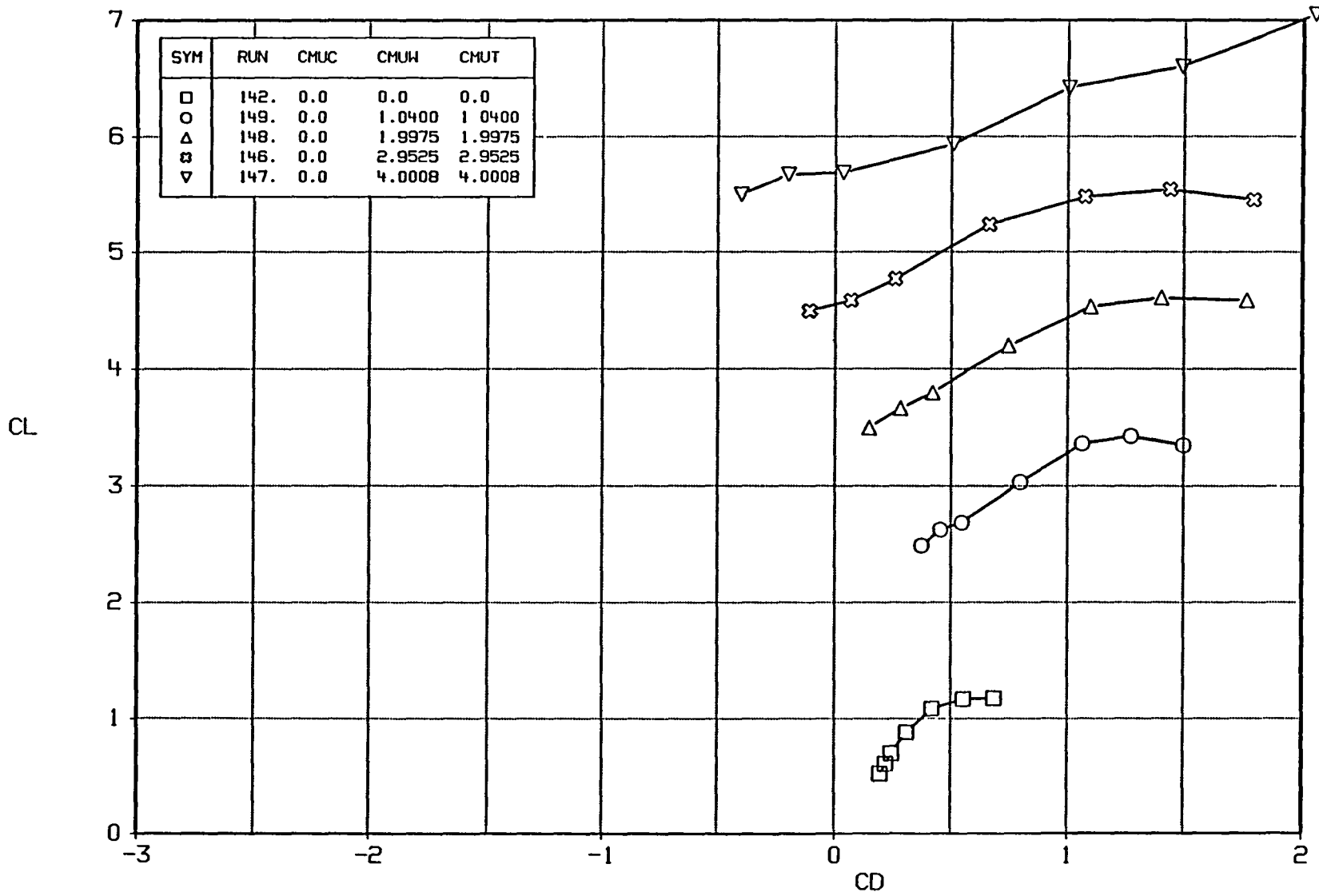


FIGURE A11b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

A-72

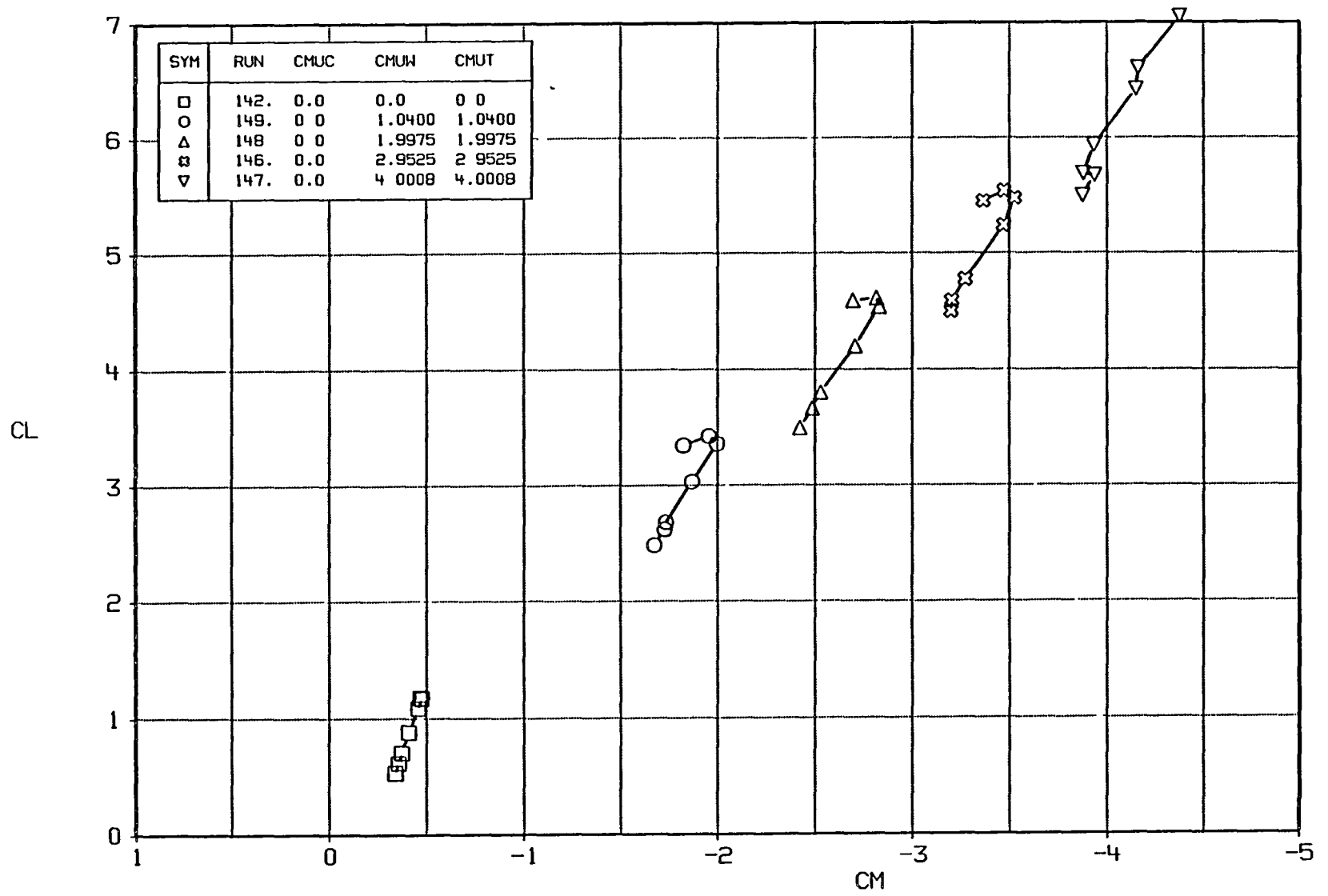


FIGURE A11c BASIC DATA EFFECT OF CMU  
CONFIGURATION BWSV, DELF=45, BN/B=1

A-73

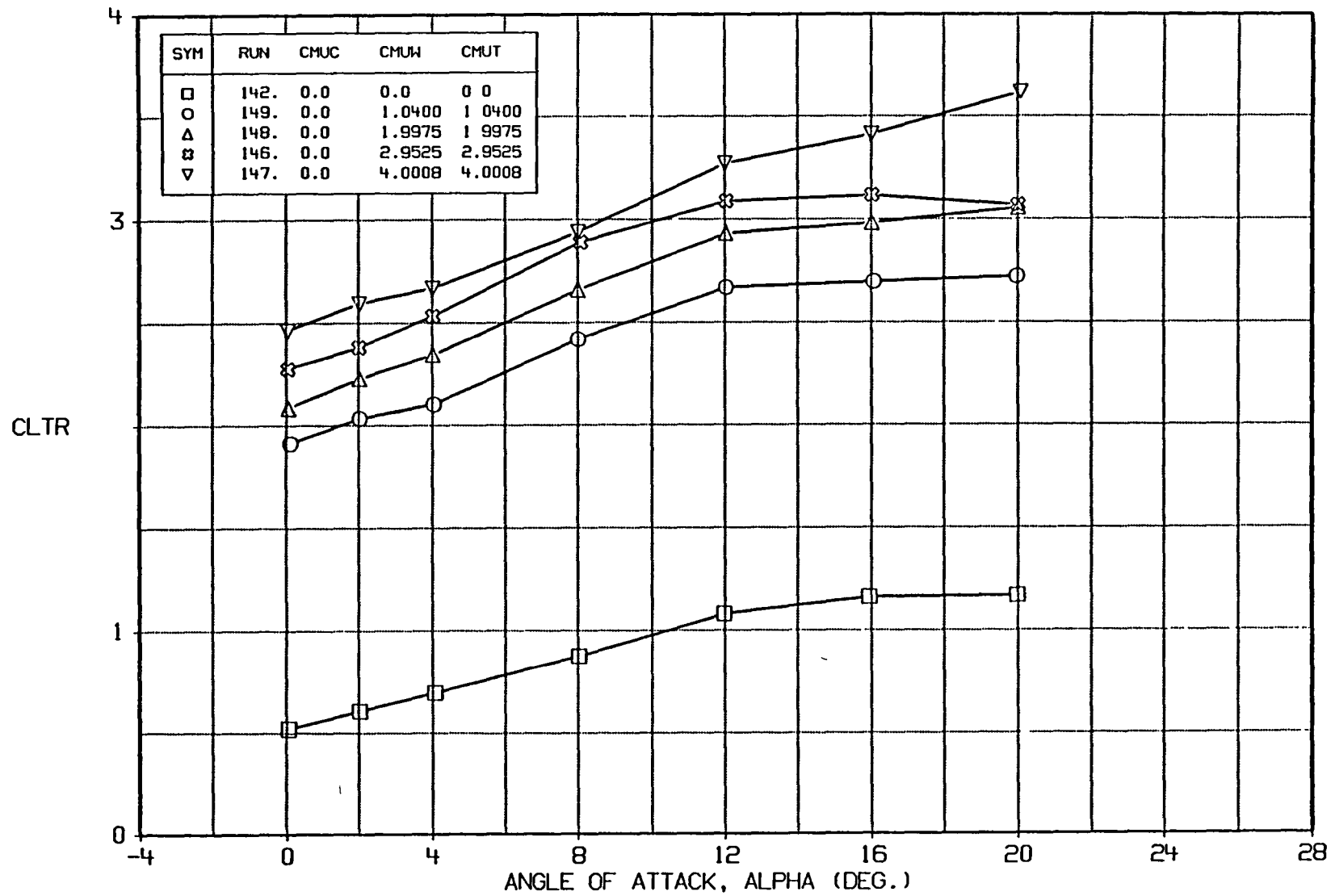


FIGURE A11d BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V, DELF=45, BN/B=1

A-74

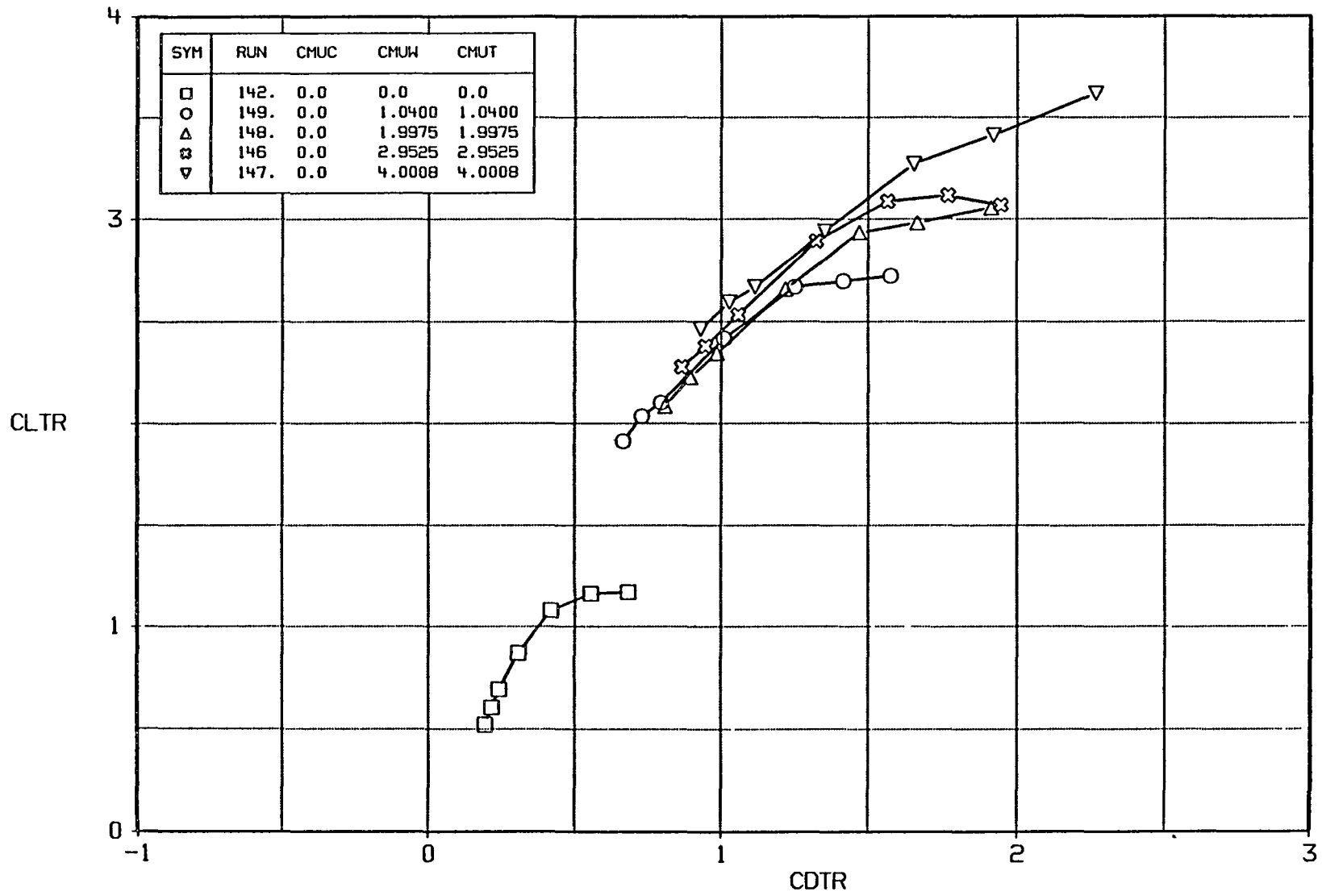


FIGURE A11e BASIC DATA EFFECT OF CMU  
CONFIGURATION BWSV, DELF=45, BN/B=1

A-75

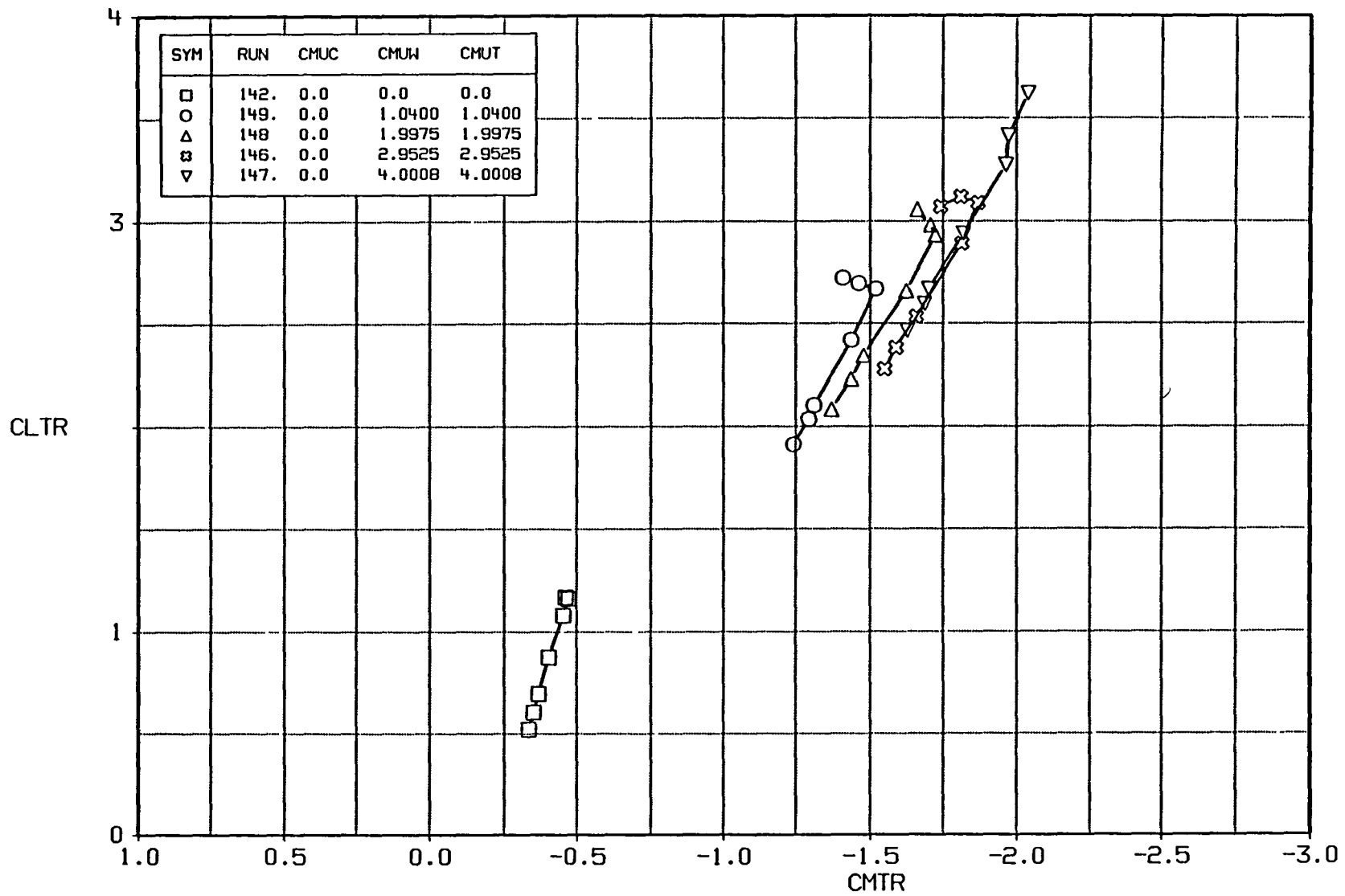


FIGURE A11f BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

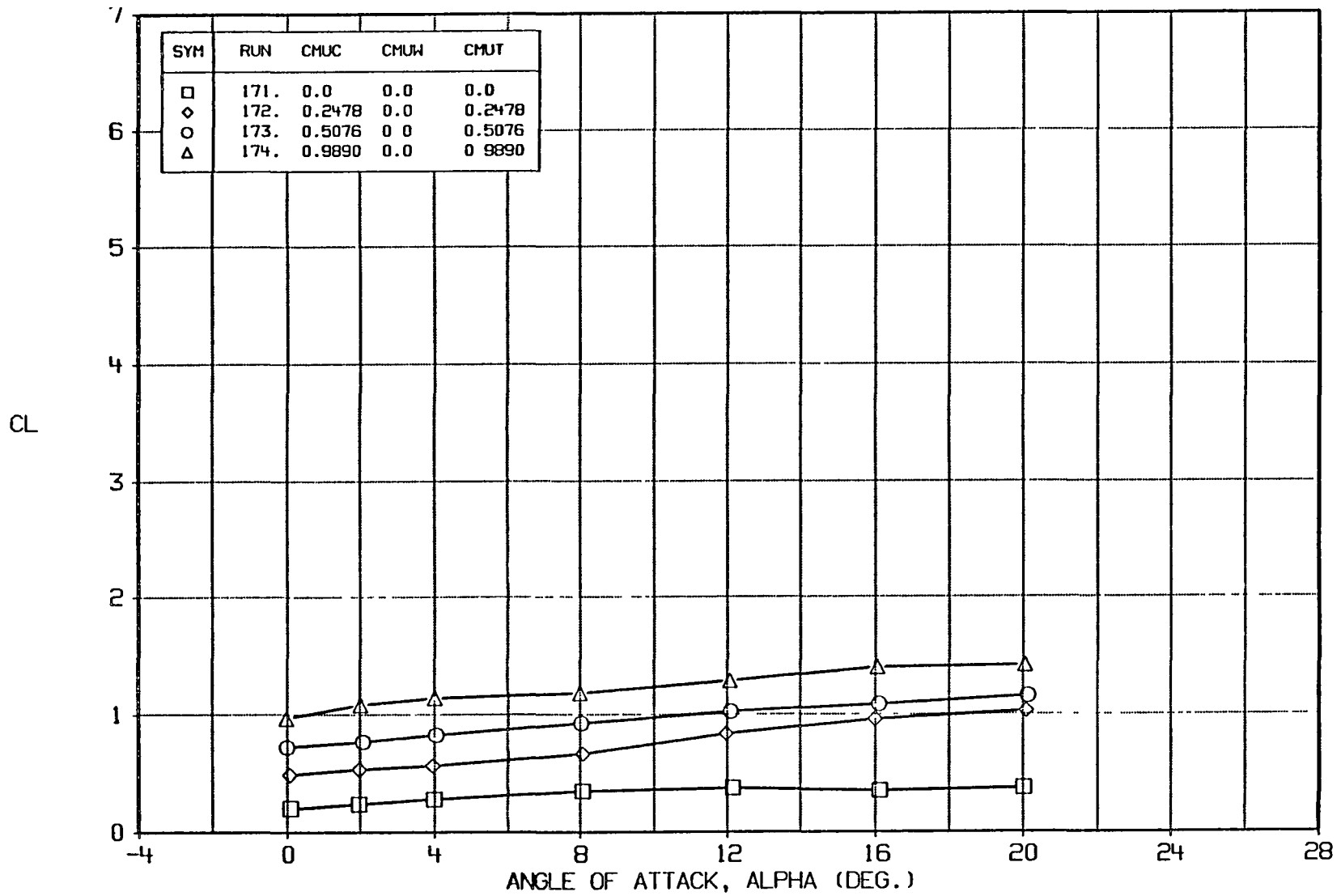


FIGURE A12a BASIC DATA EFFECT OF CMU  
 CONFIGURATION BCIV, DELF=45, DELC=10, BN/B=1

A-77

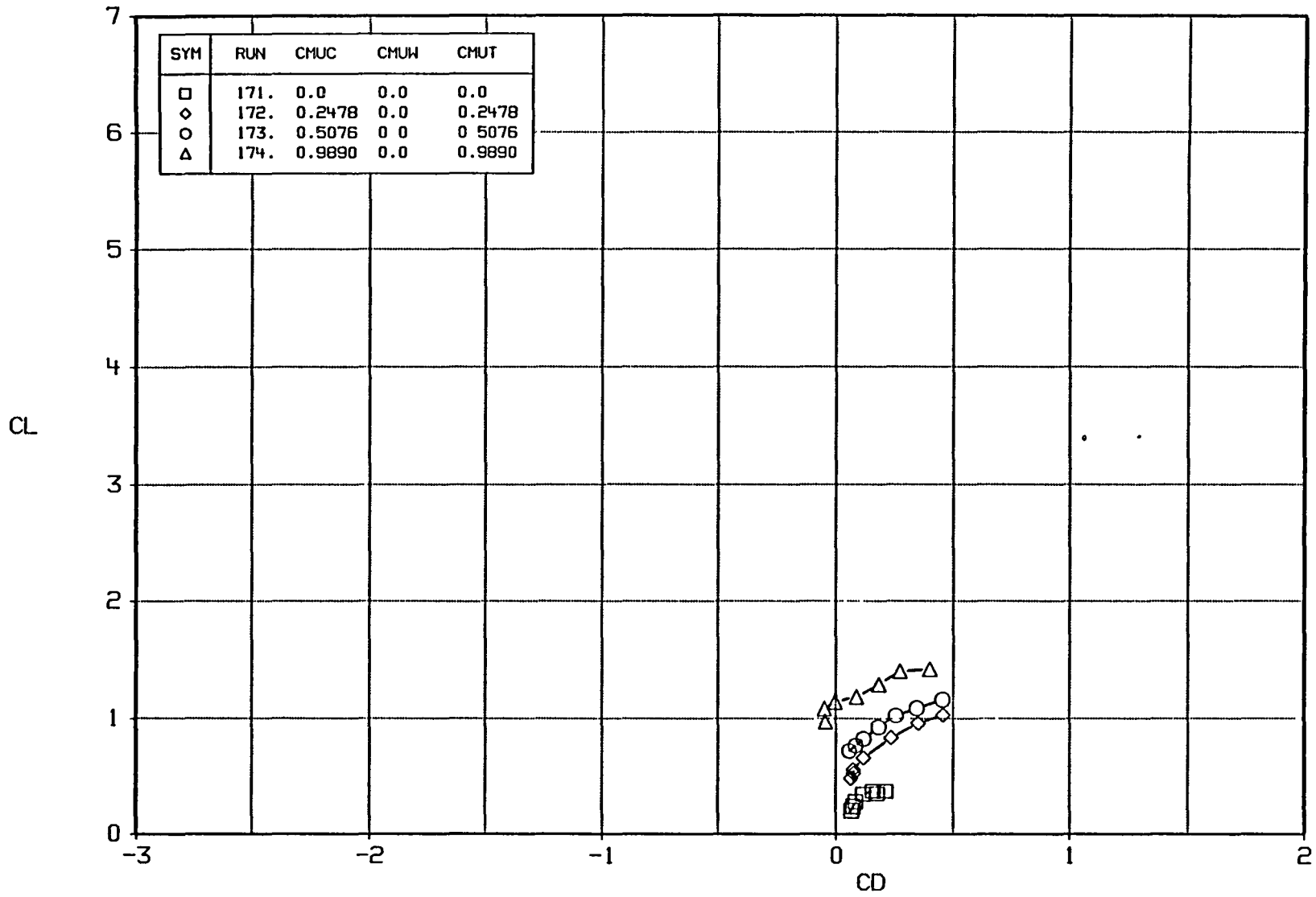


FIGURE A12b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, DELC=10, BN/B=1

A-78

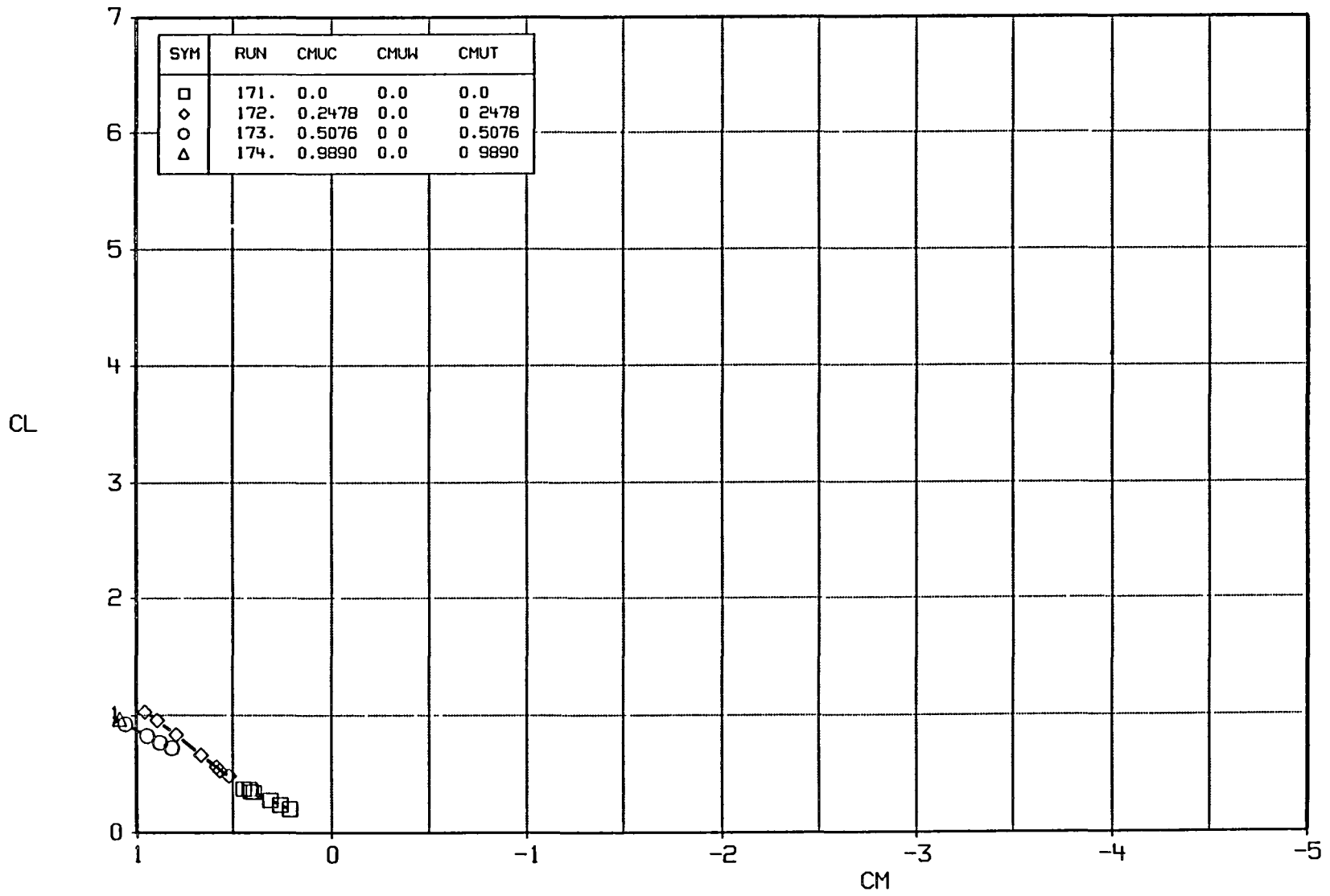


FIGURE A12c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, DELC=10, BN/B=1



A-79

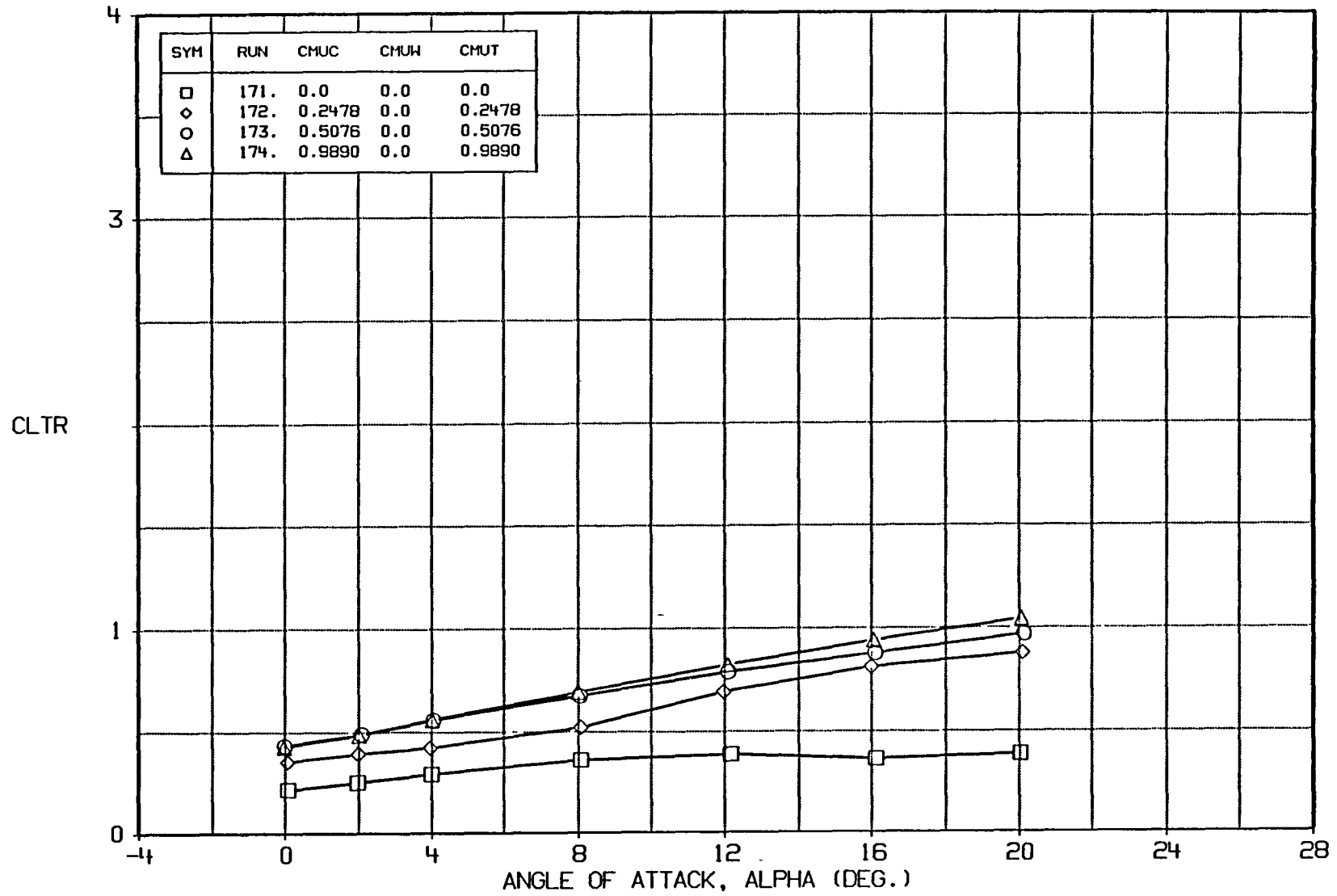


FIGURE A12d BASIC DATA EFFECT OF CMU  
CONFIGURATION BCIV, DELF=45, DELC=10, BN/B=1

A-80

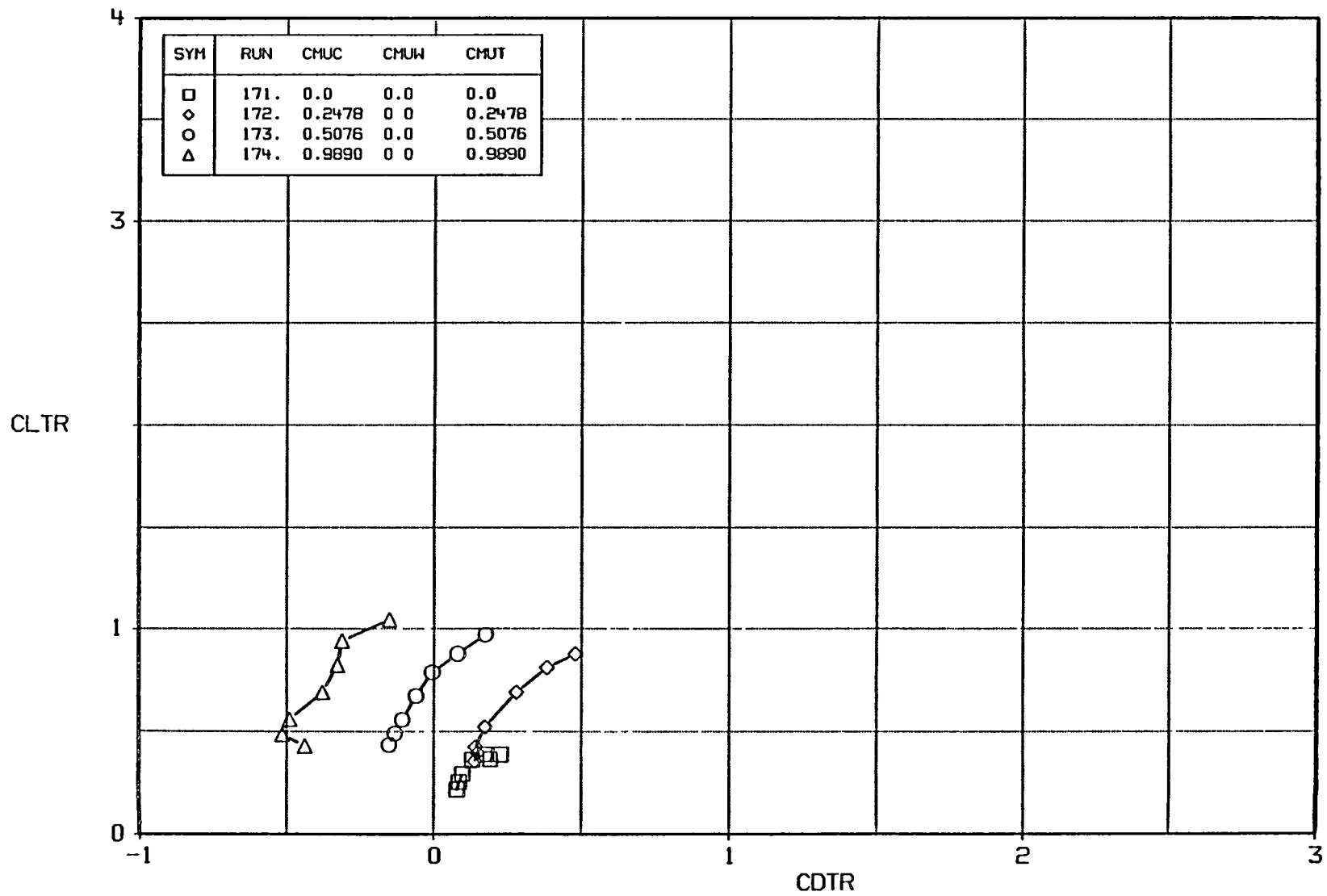


FIGURE A12e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, DELC=10, BN/B=1

A-81

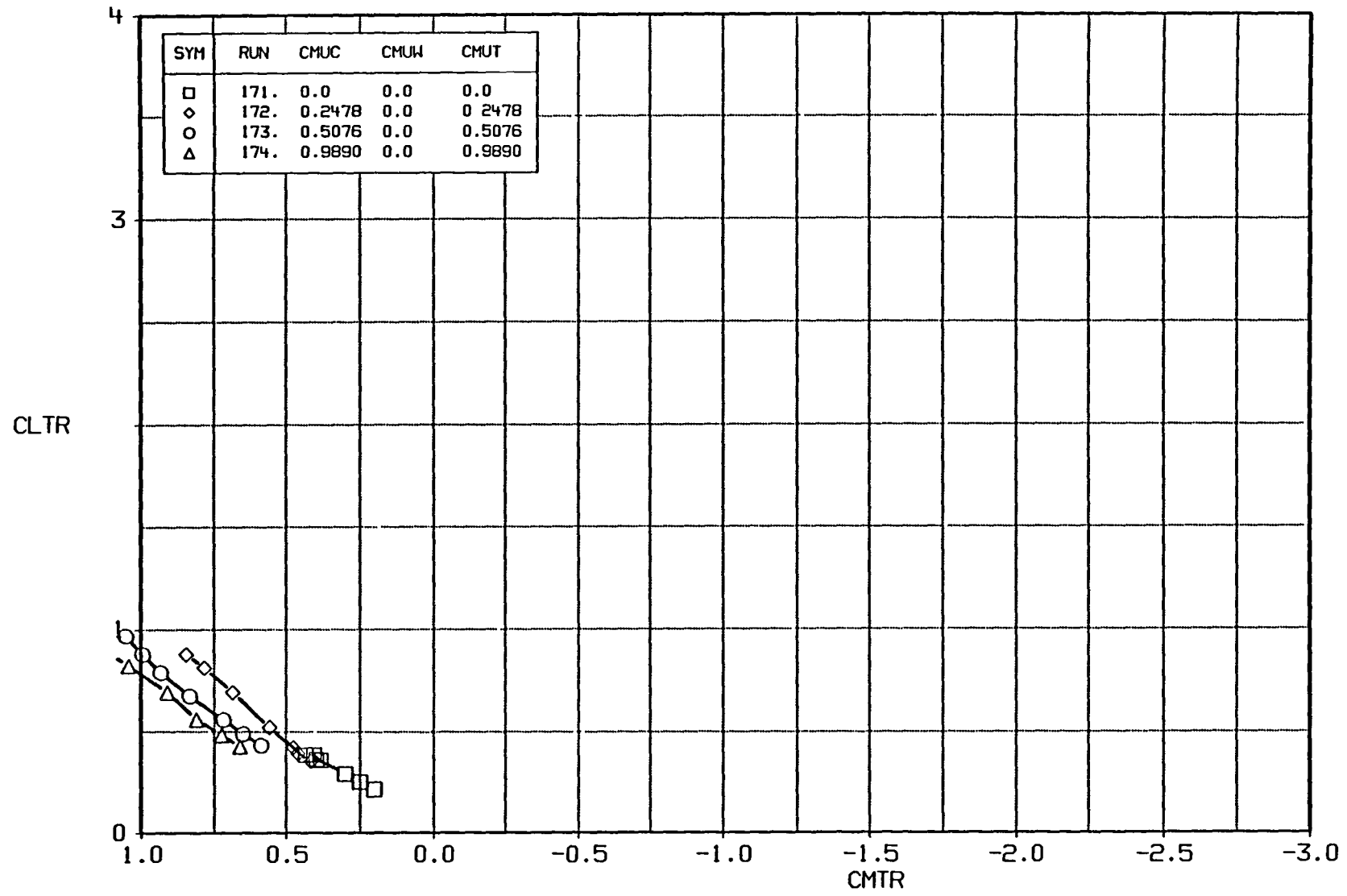


FIGURE A12f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, DELC=10, BN/B=1

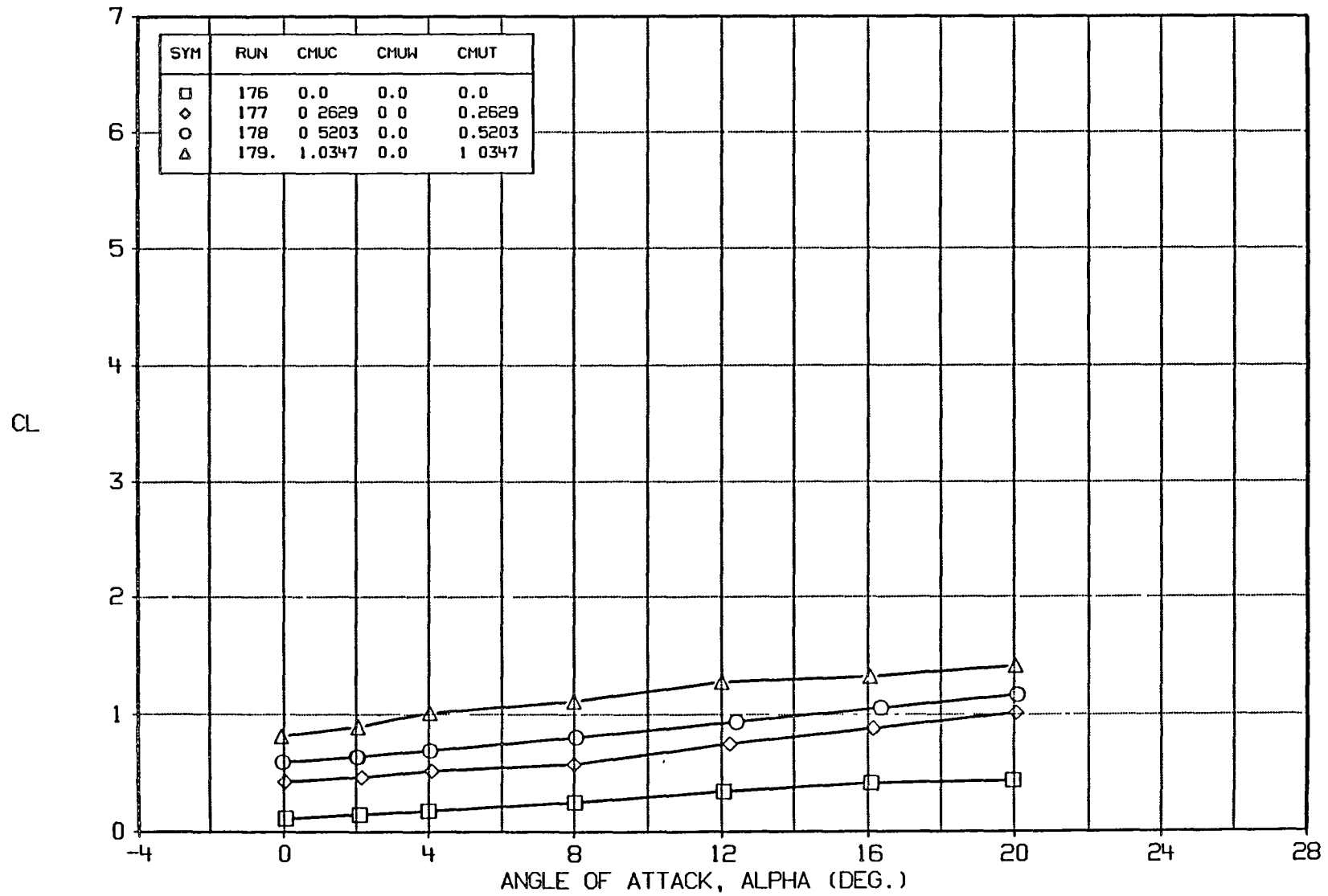


FIGURE A13a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, BN/B=1

A-83

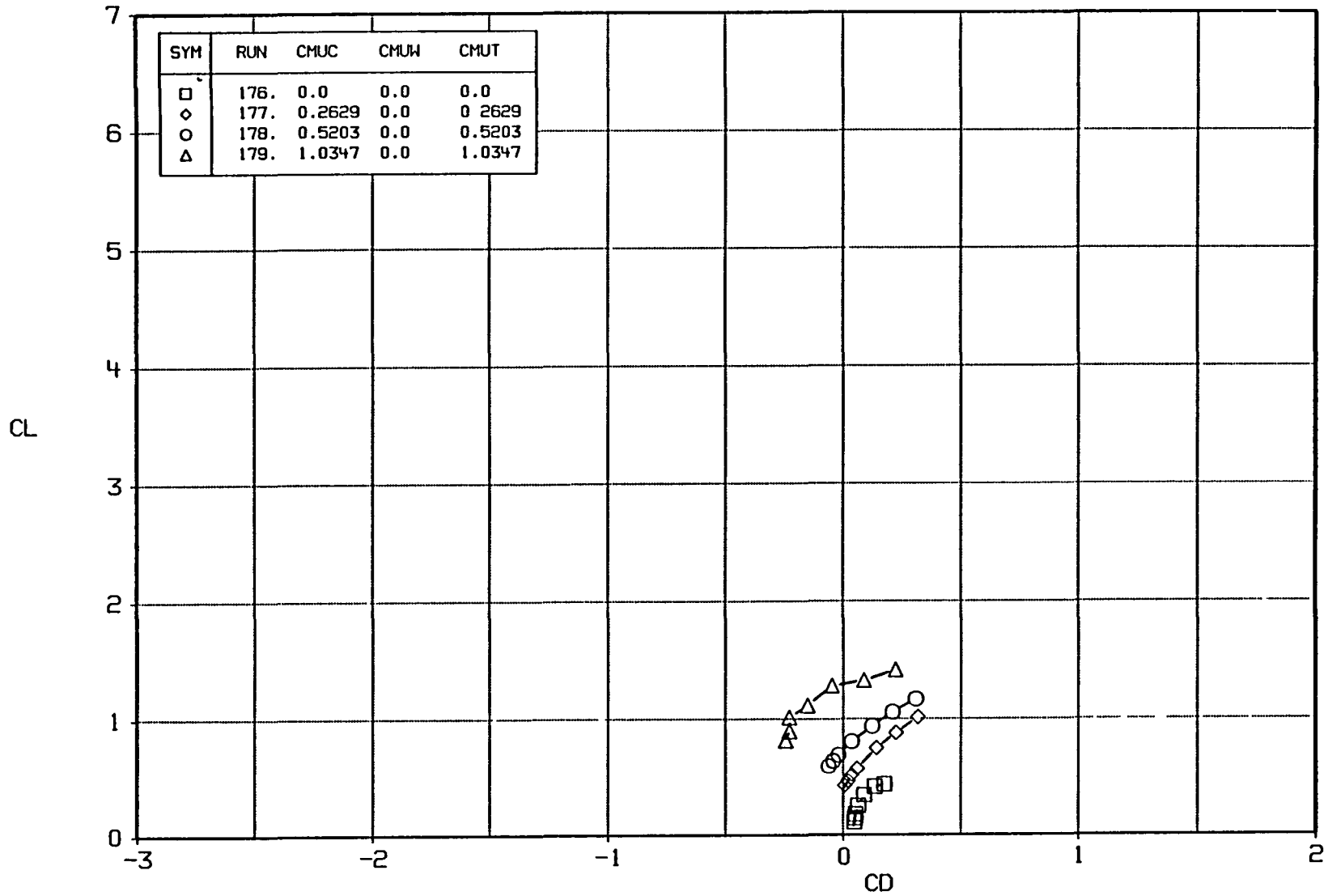


FIGURE A13b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, BN/B=1

A-84

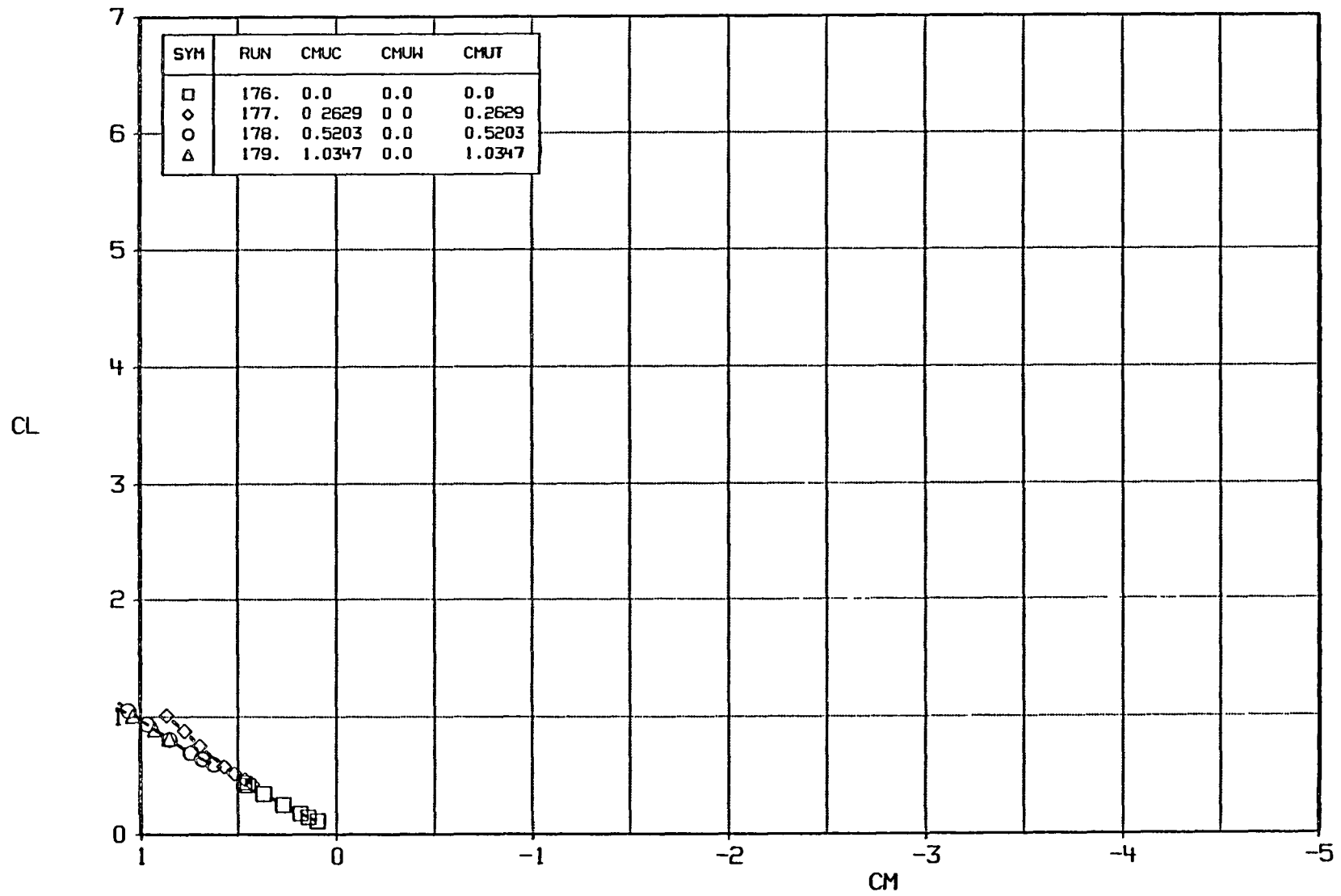


FIGURE A13c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, BN/B=1

A-85

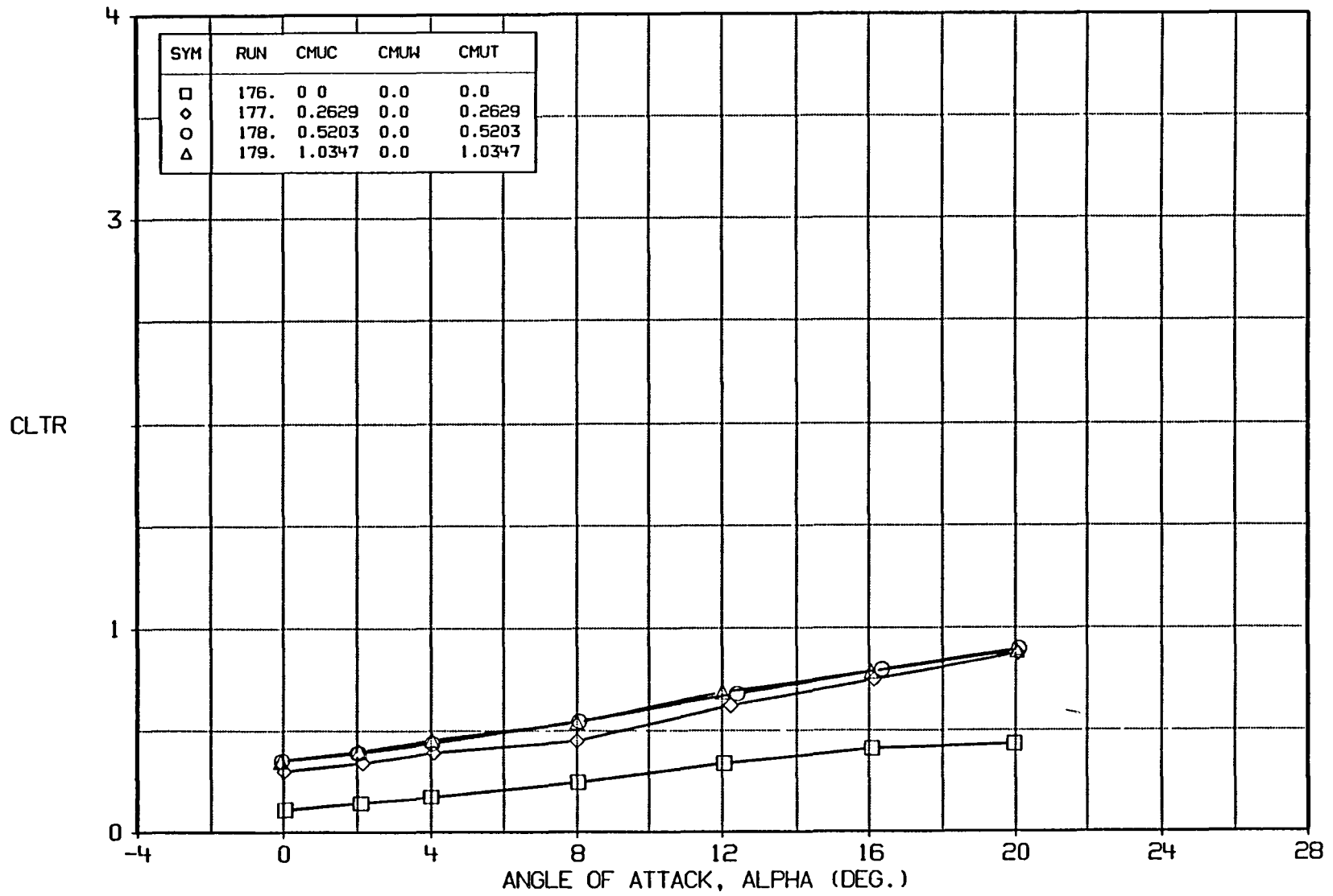


FIGURE A13d BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, BN/B=1

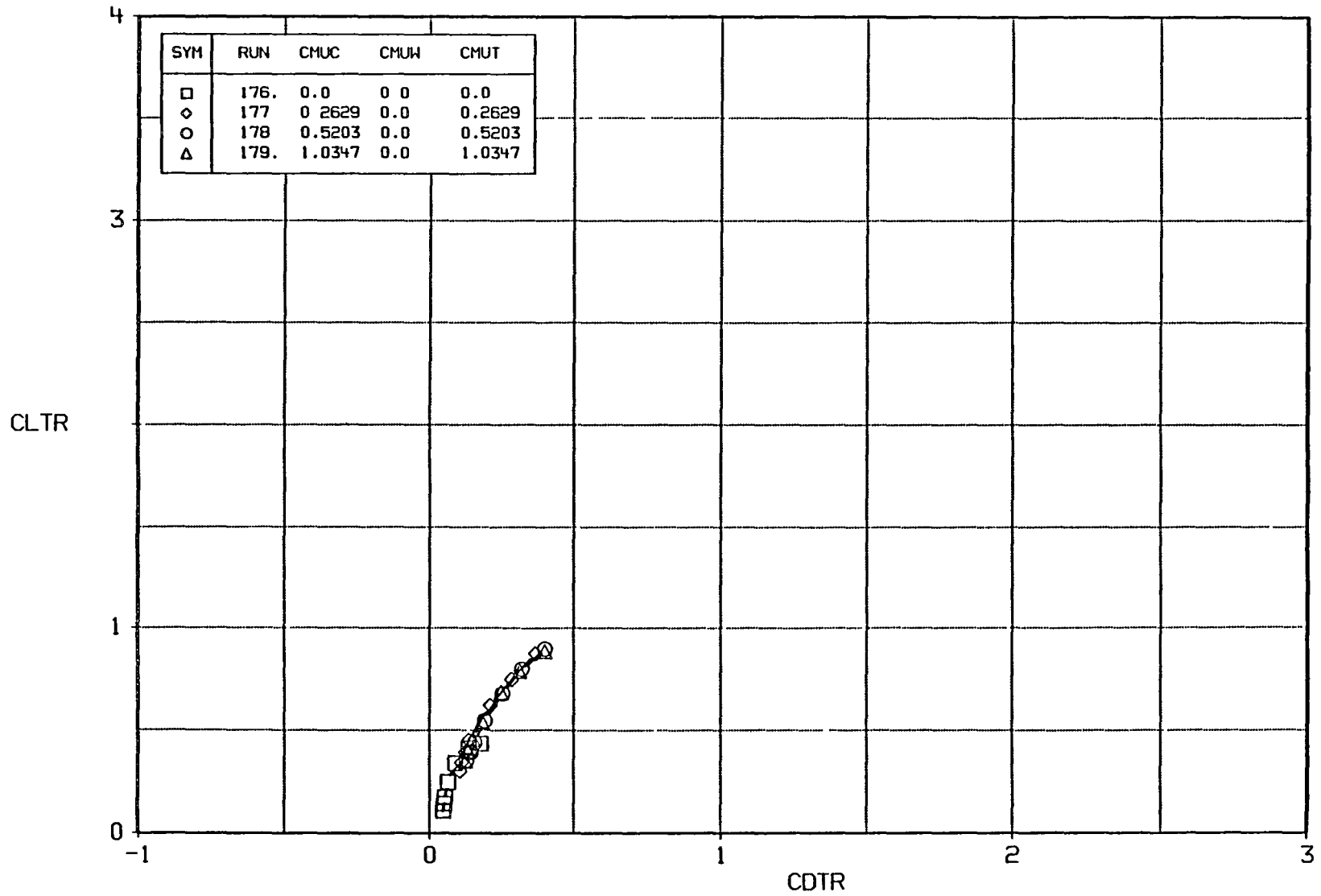


FIGURE A13e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, BN/B=1



A-87

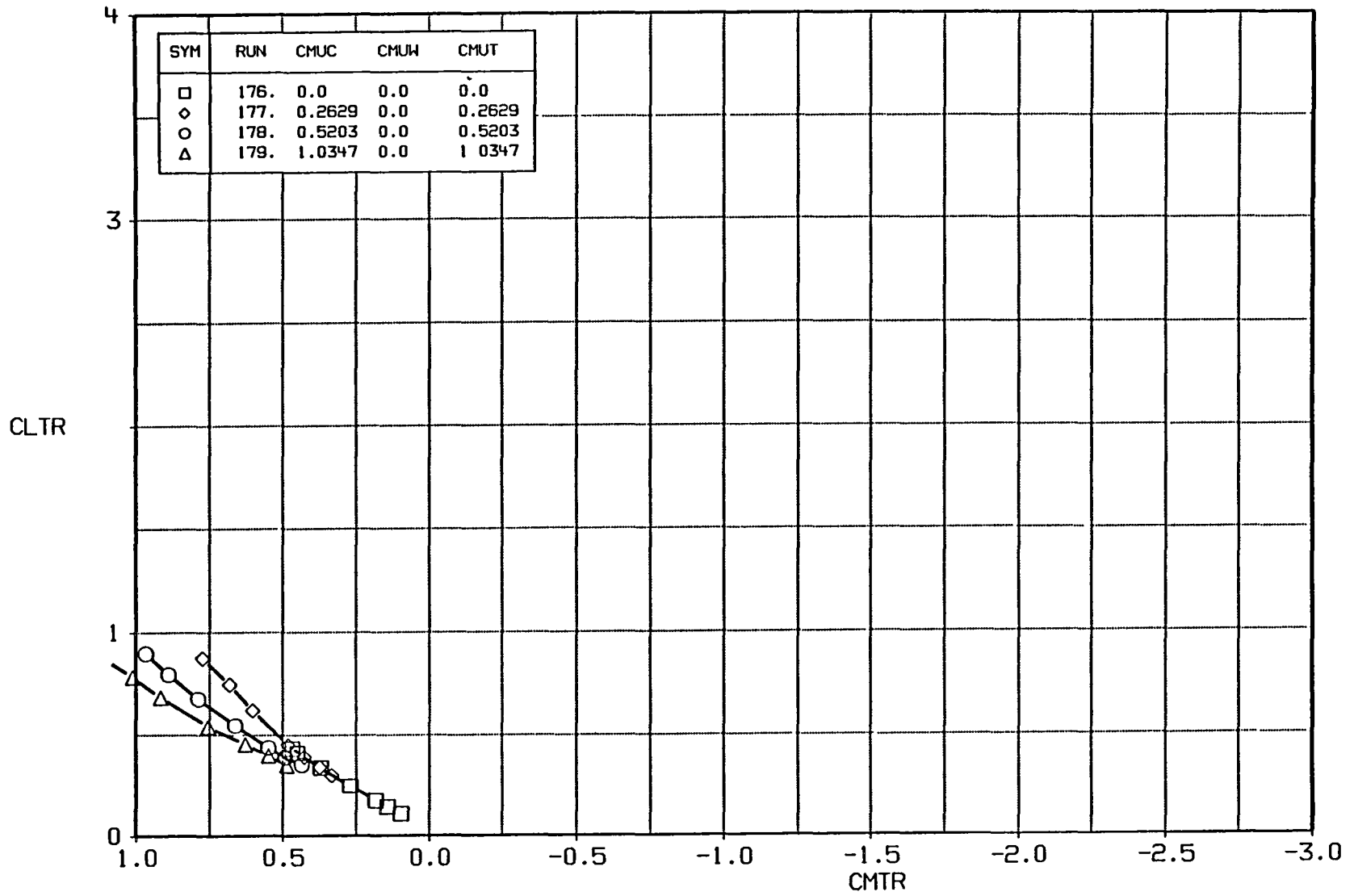


FIGURE A13f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1V, DELF=45, BN/B=1

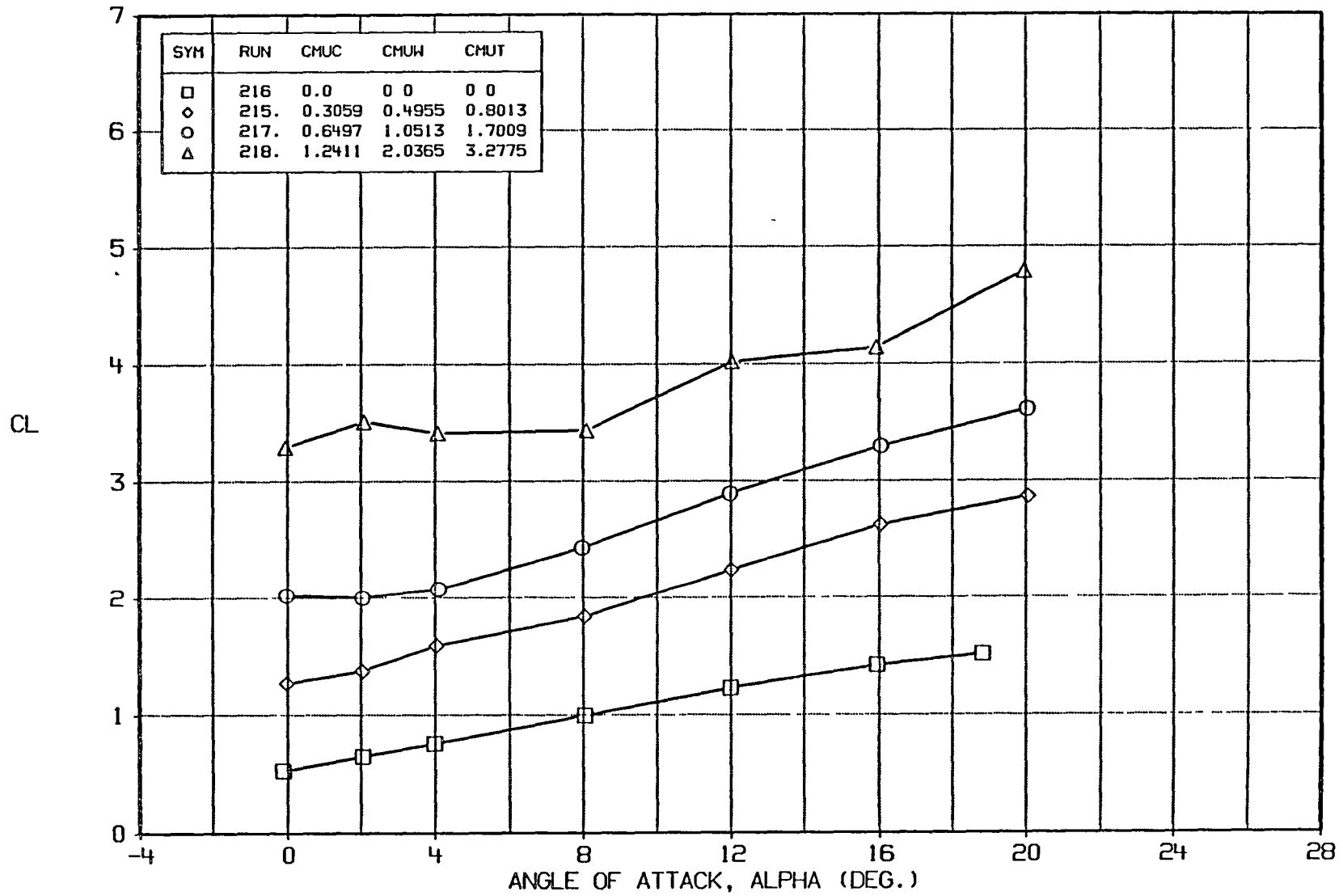


FIGURE A14a BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=45

A-89

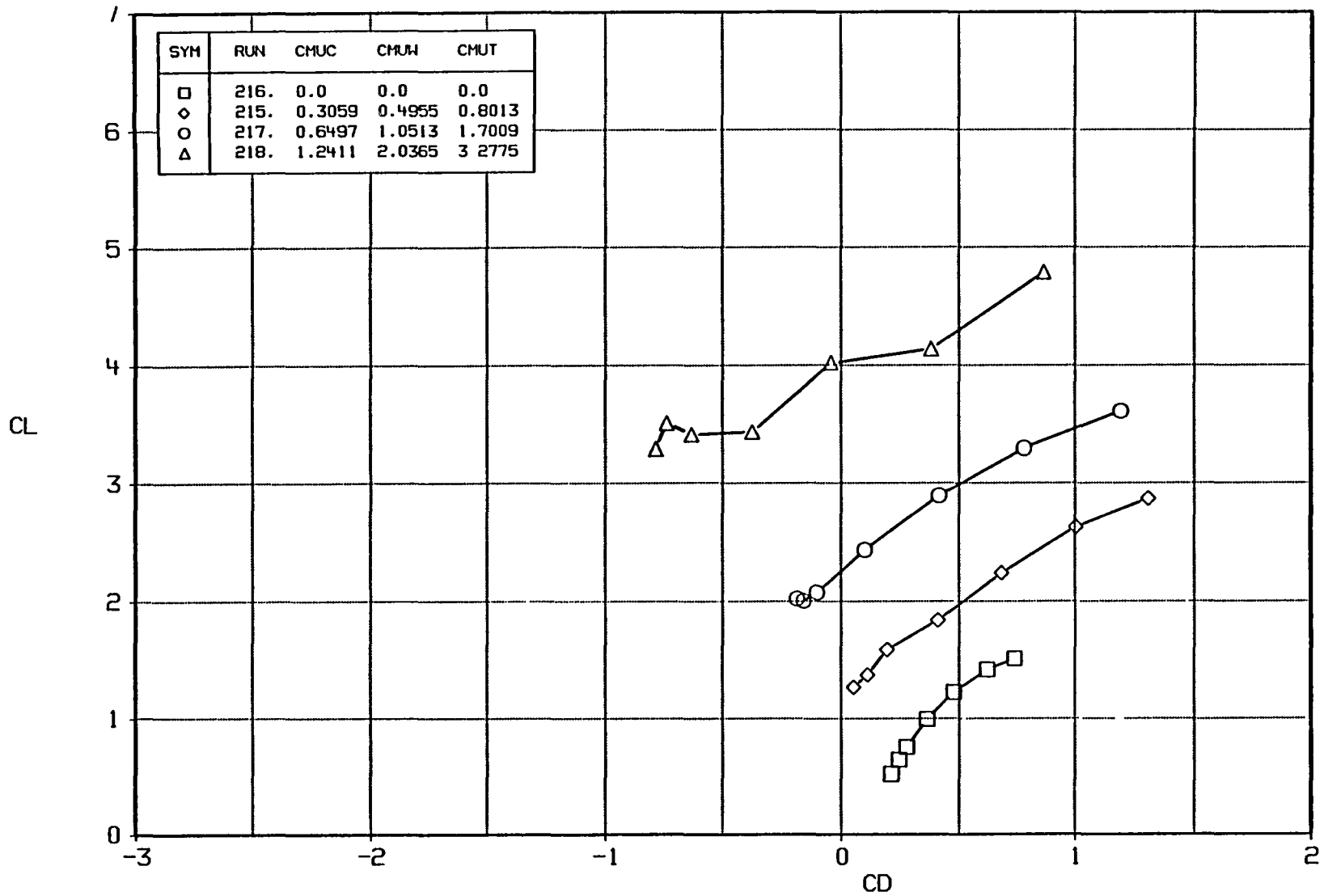


FIGURE A14b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=45

A-90

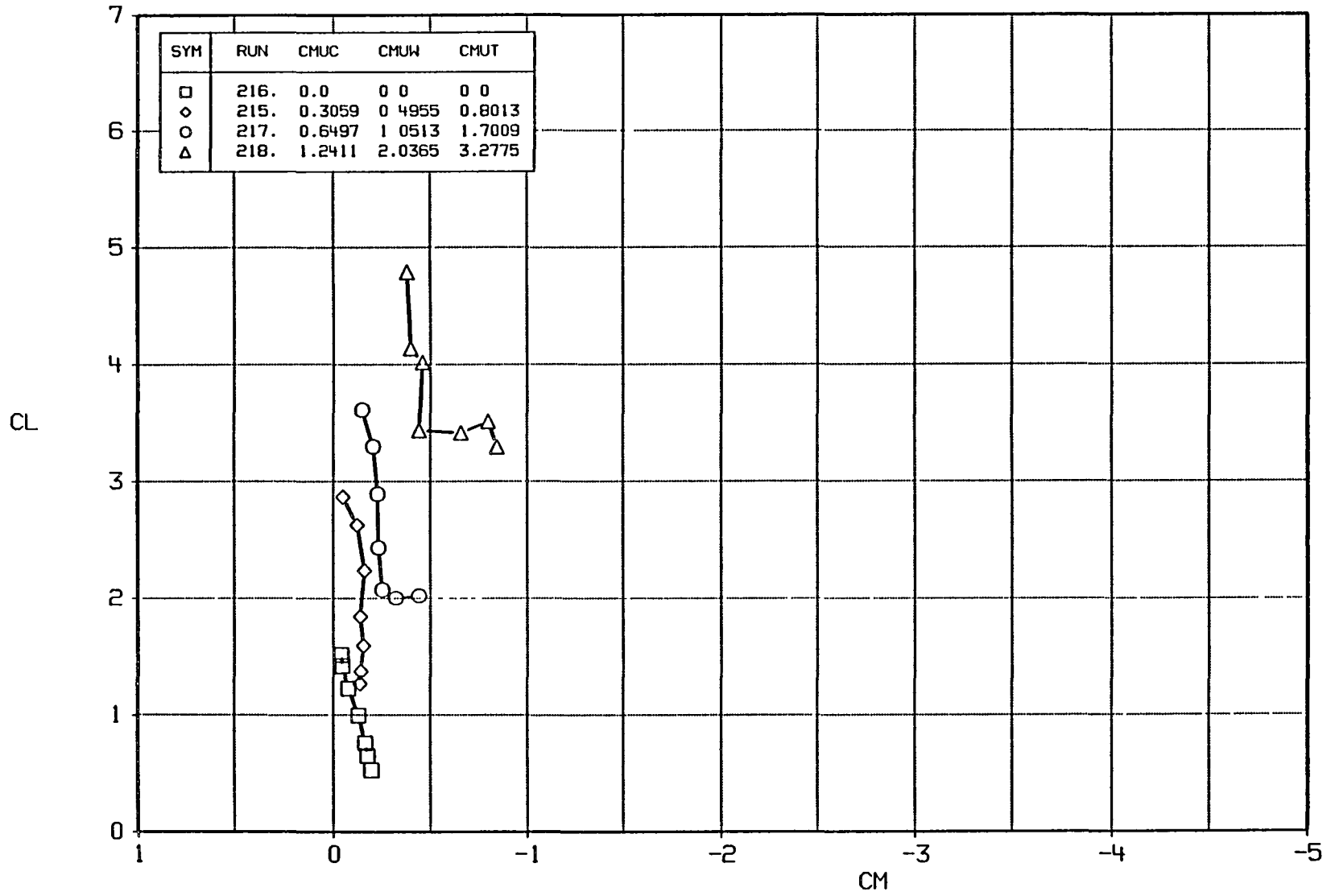


FIGURE A14c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=45

A-91

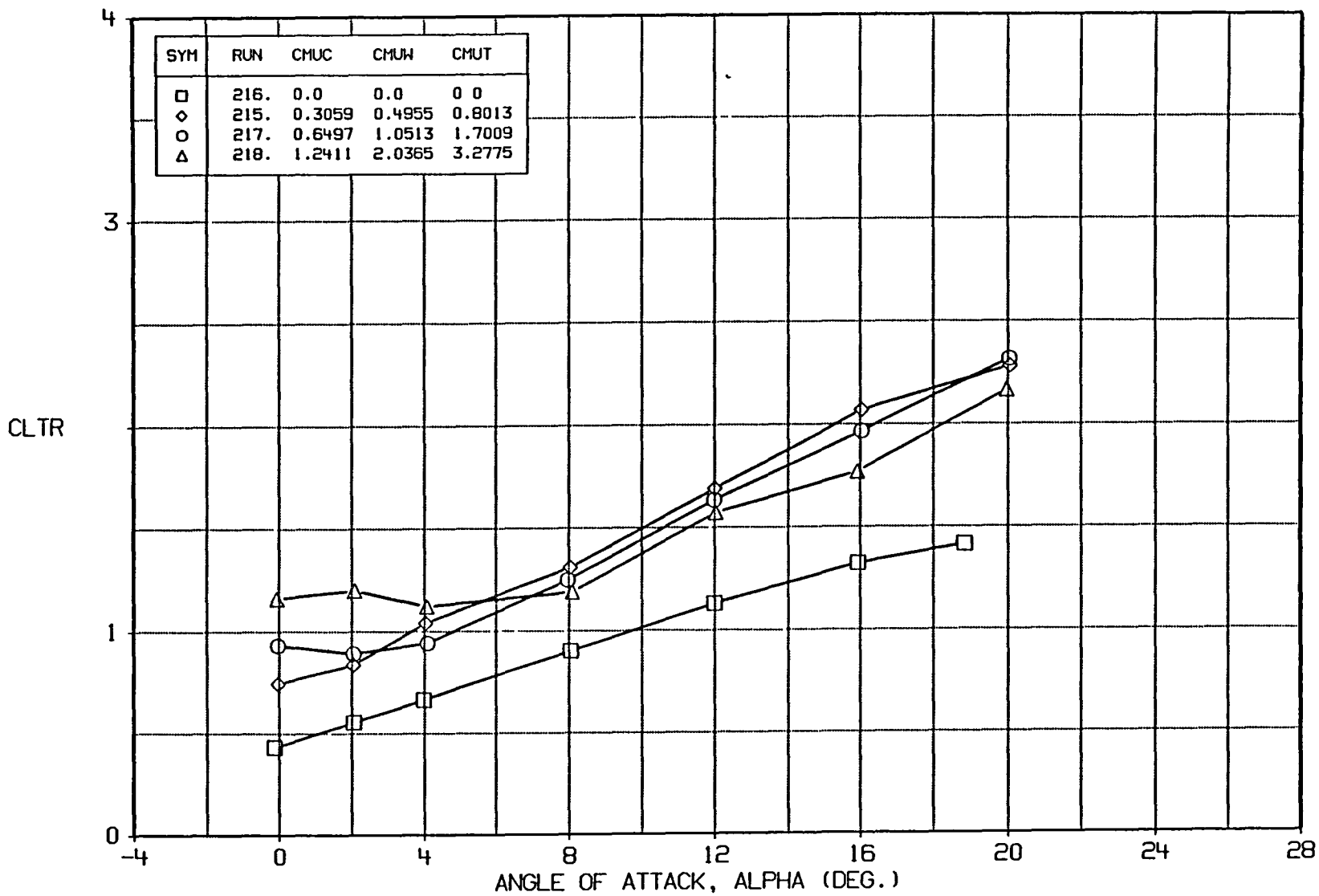


FIGURE A14d BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=45

A-92

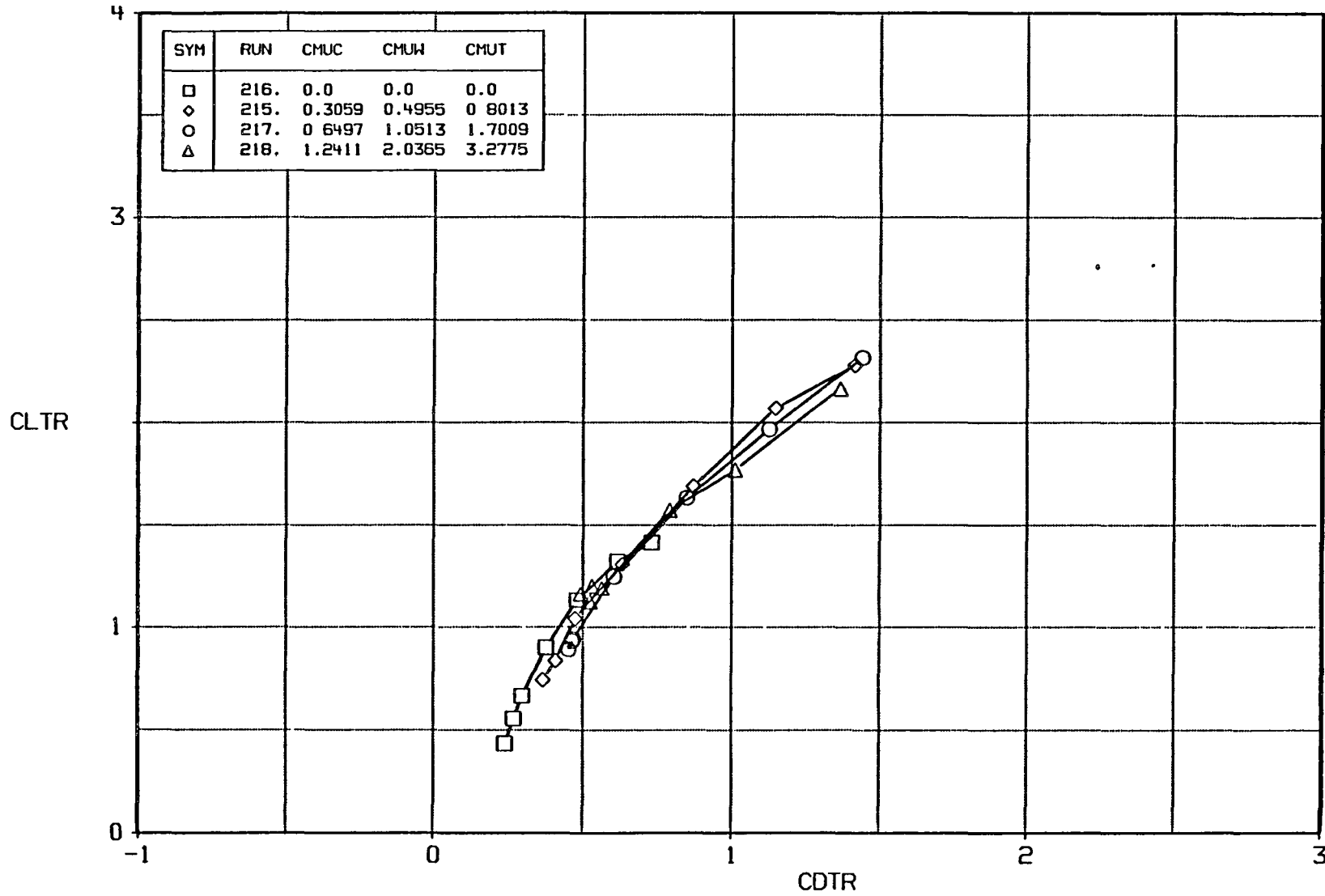


FIGURE A14e BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=45

A-93

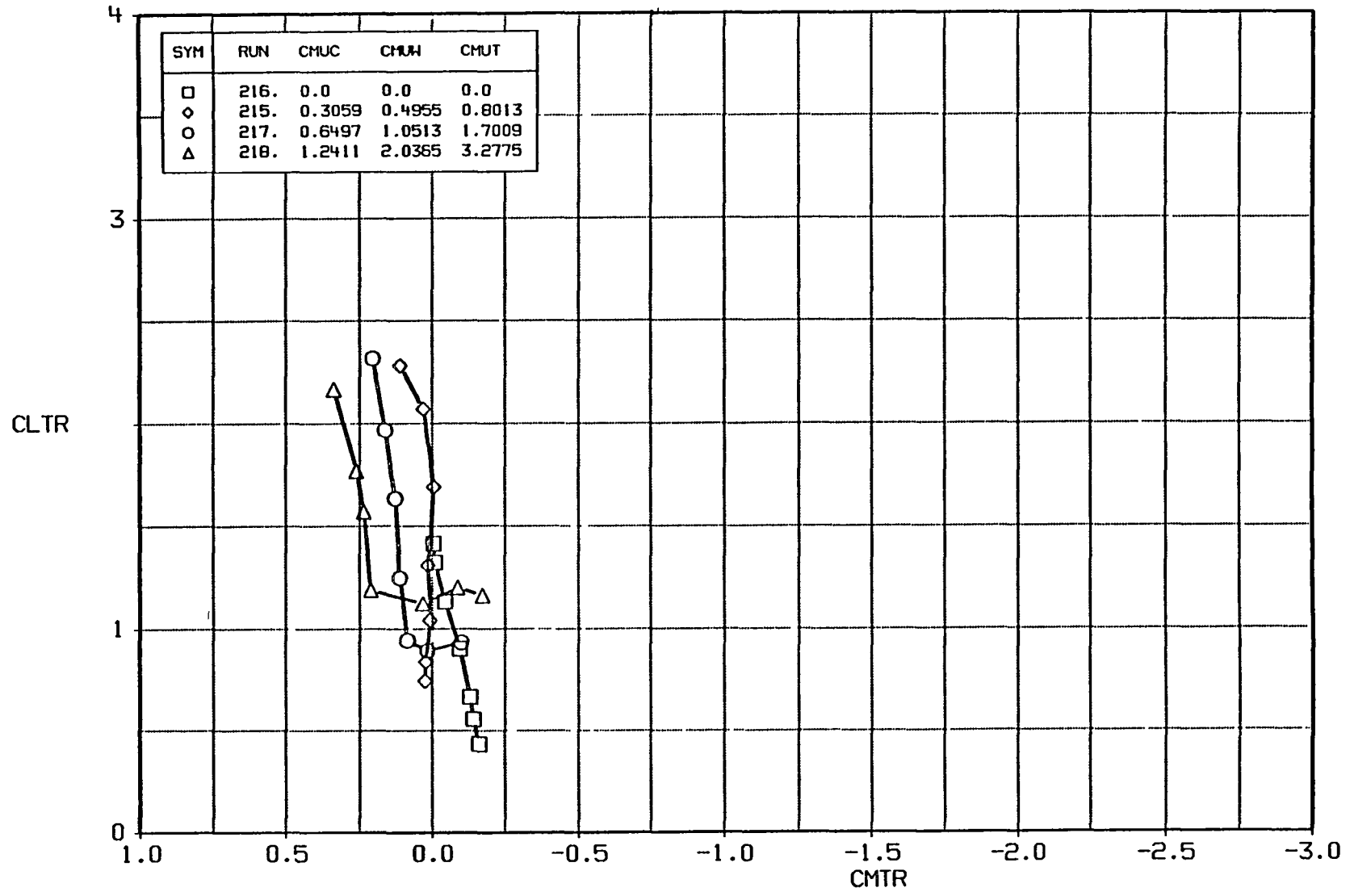


FIGURE A14f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=45

A-94

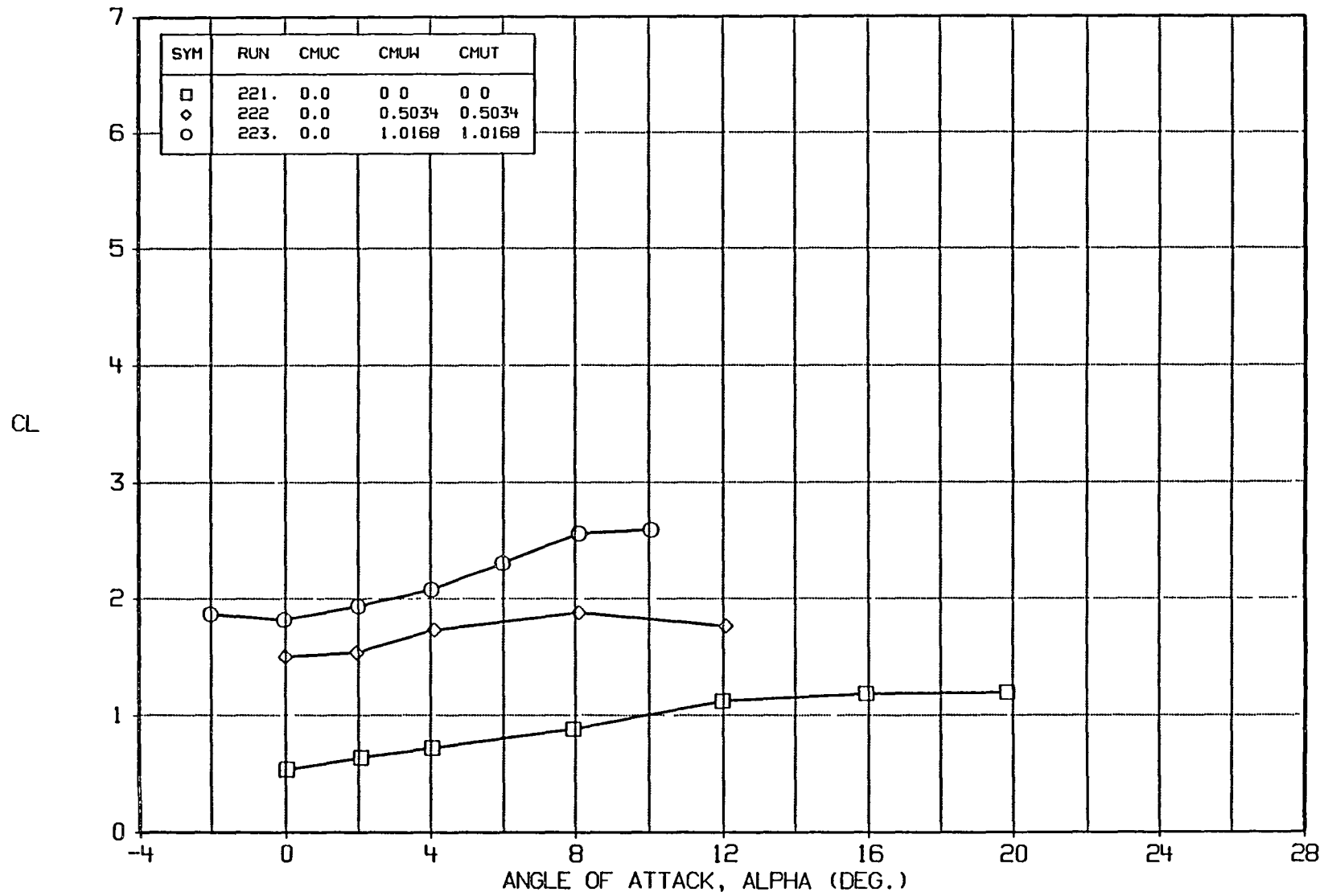


FIGURE A15a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5, DELF=45



A-95

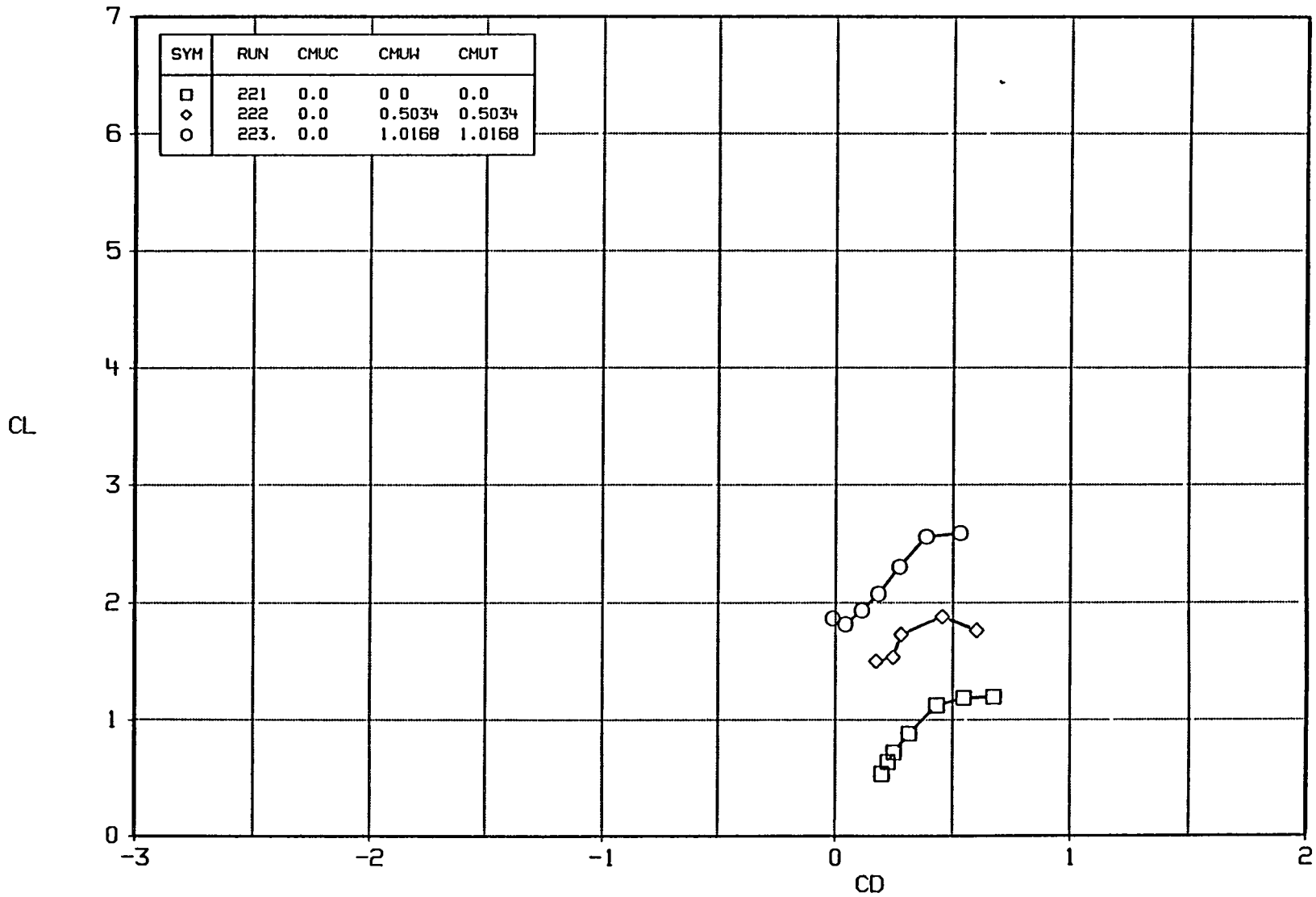


FIGURE A15b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5, DELF=45

A-96

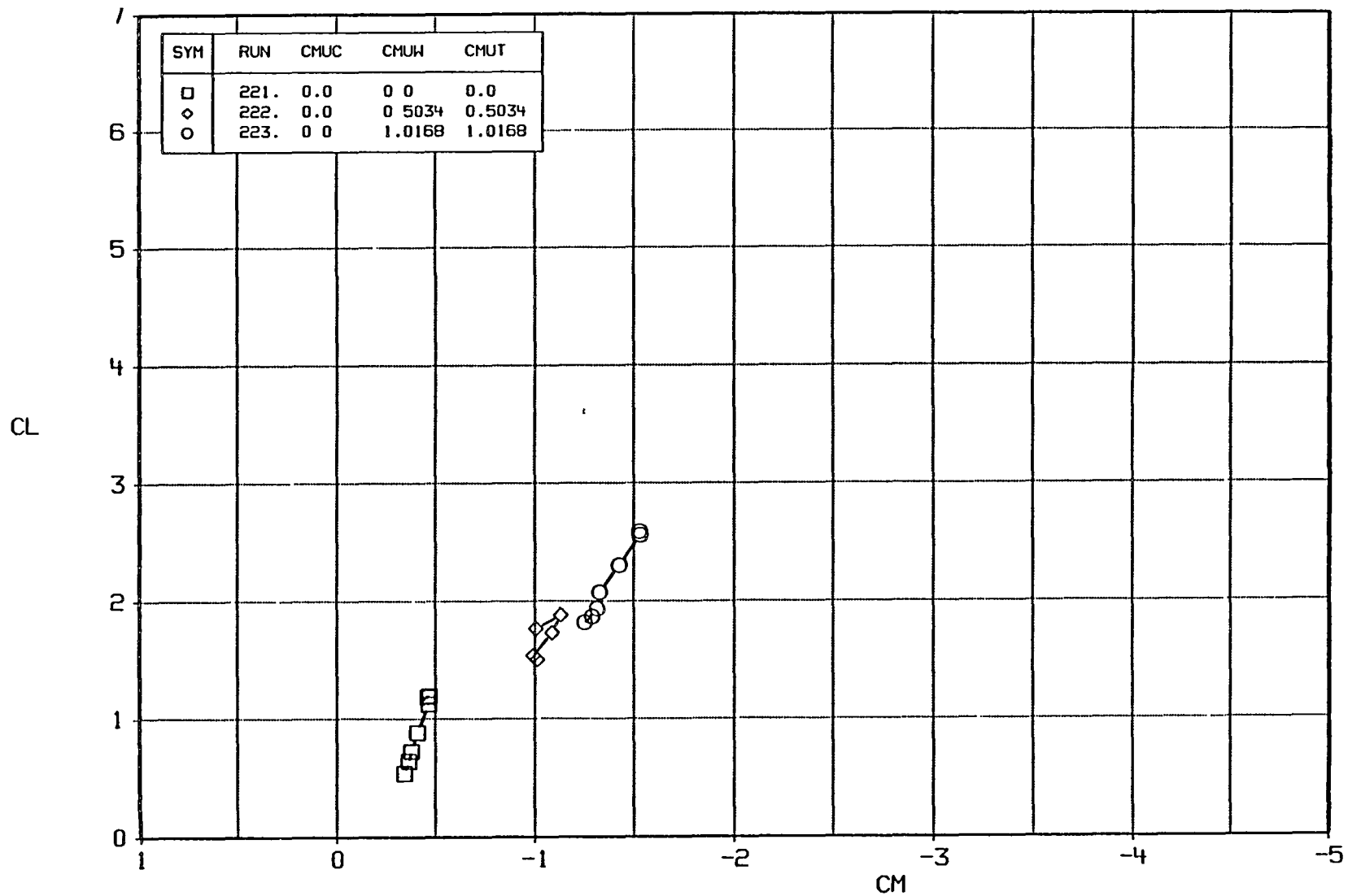


FIGURE A15c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5, DELF=45

A-97

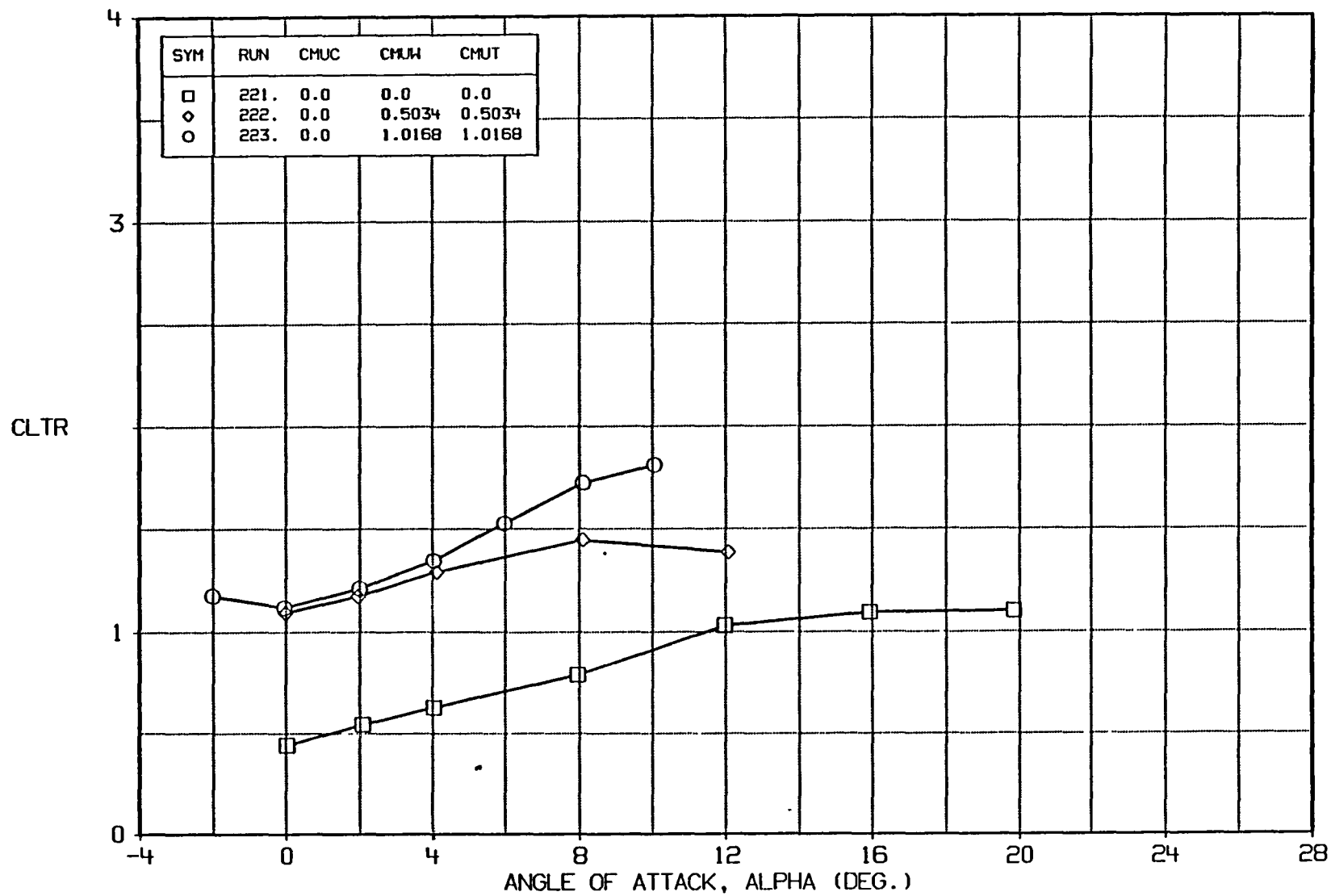


FIGURE A15d BASIC DATA EFFECT OF CMU  
CONFIGURATION BWGV, BN/B=0.5, DELF=45

A-98

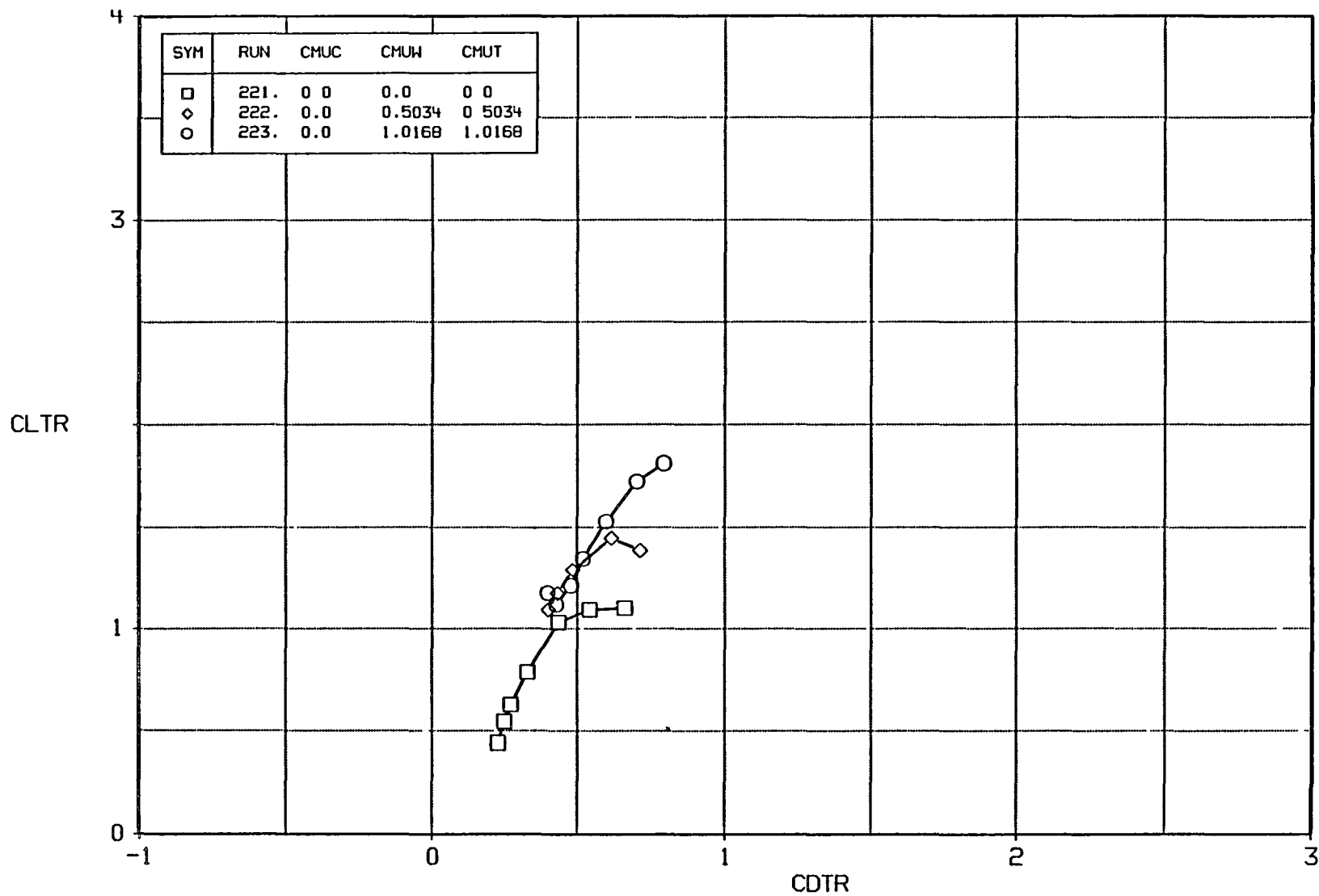


FIGURE A15e BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5, DELF=45

A-99

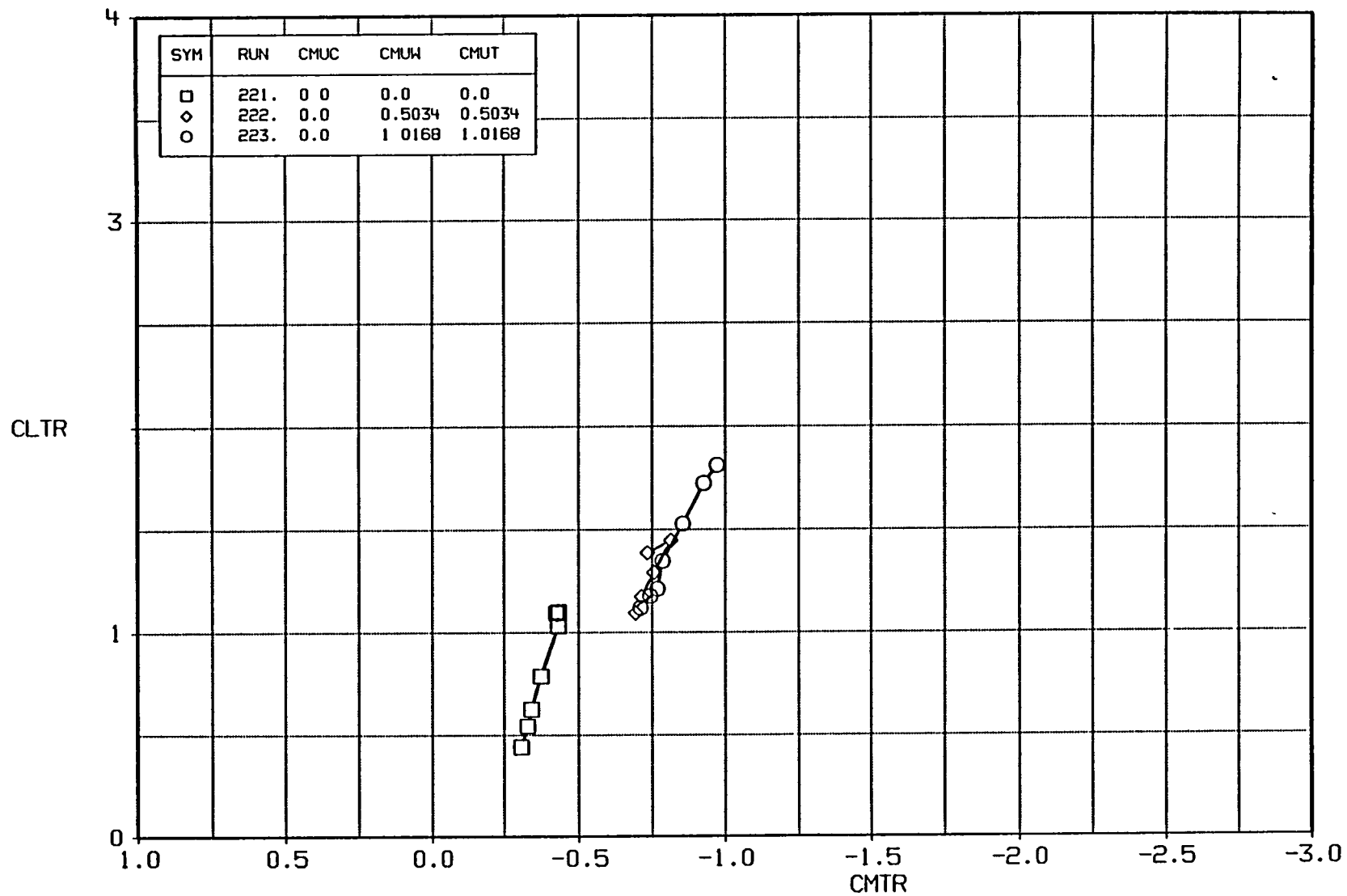


FIGURE A15f BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5, DELF=45

A-100

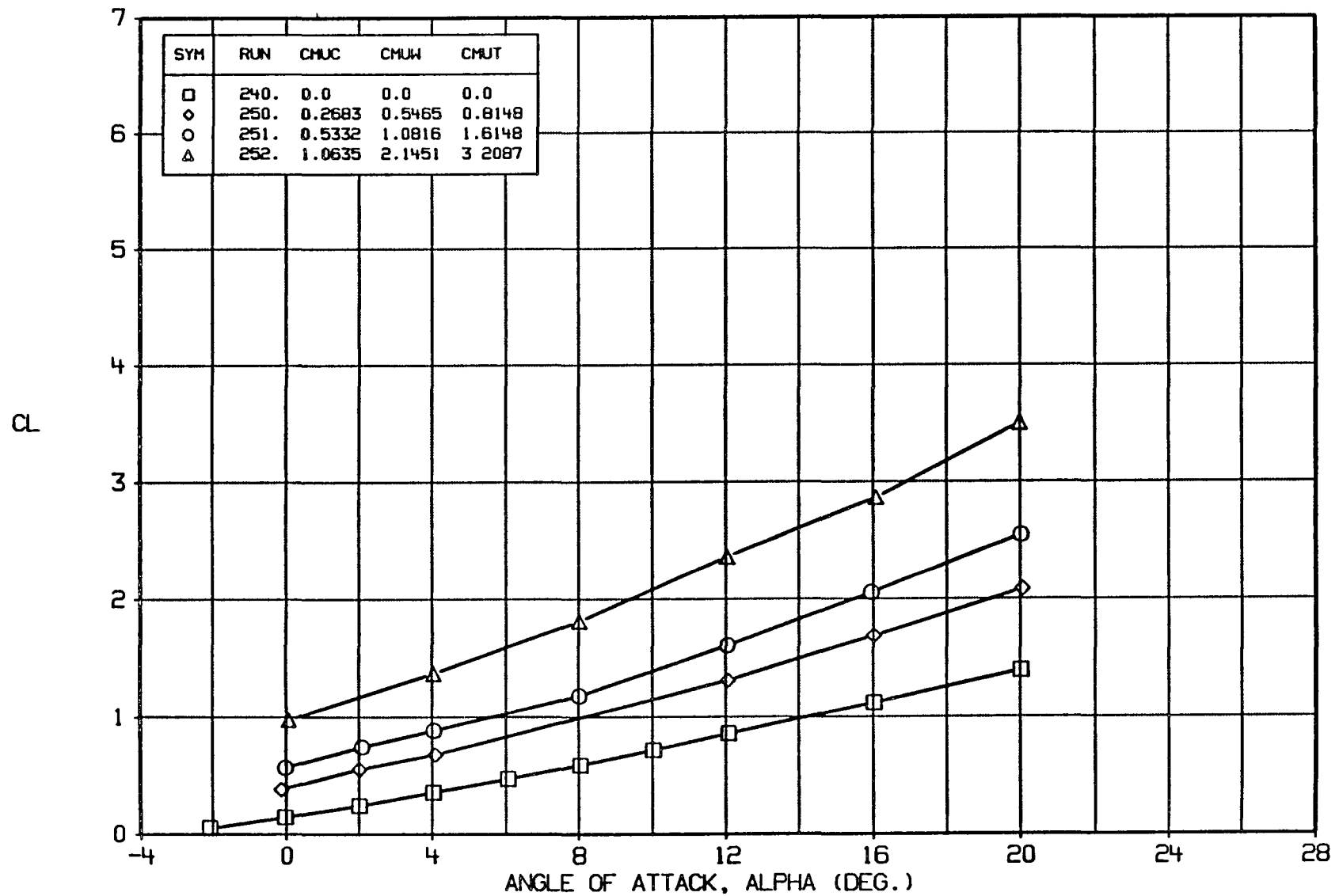


FIGURE A16a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V,  $(BN/B)C=1$ ,  $(BN/B)W=0.5$ ,  $DEL F=0$

A-101

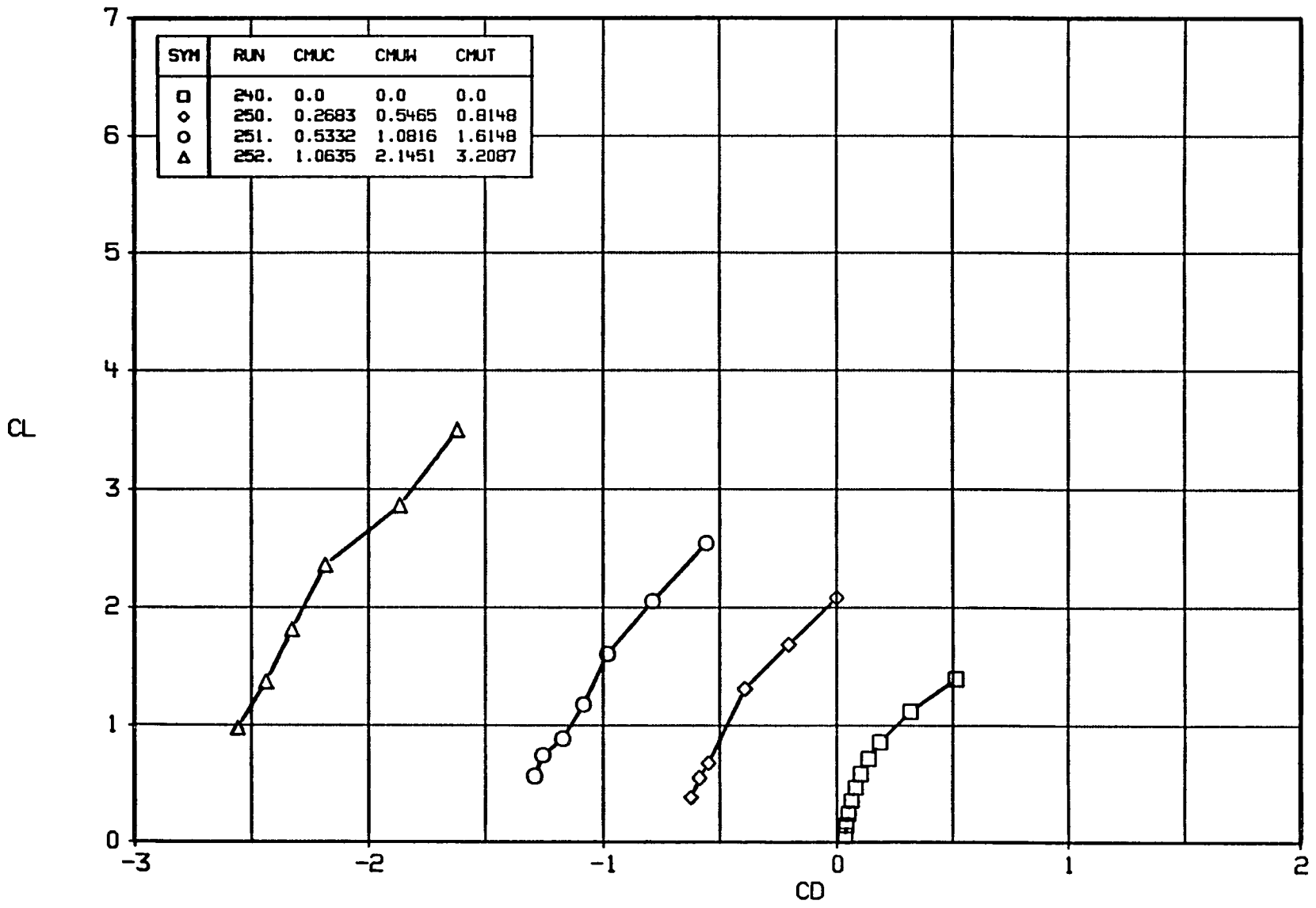


FIGURE A16b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=0

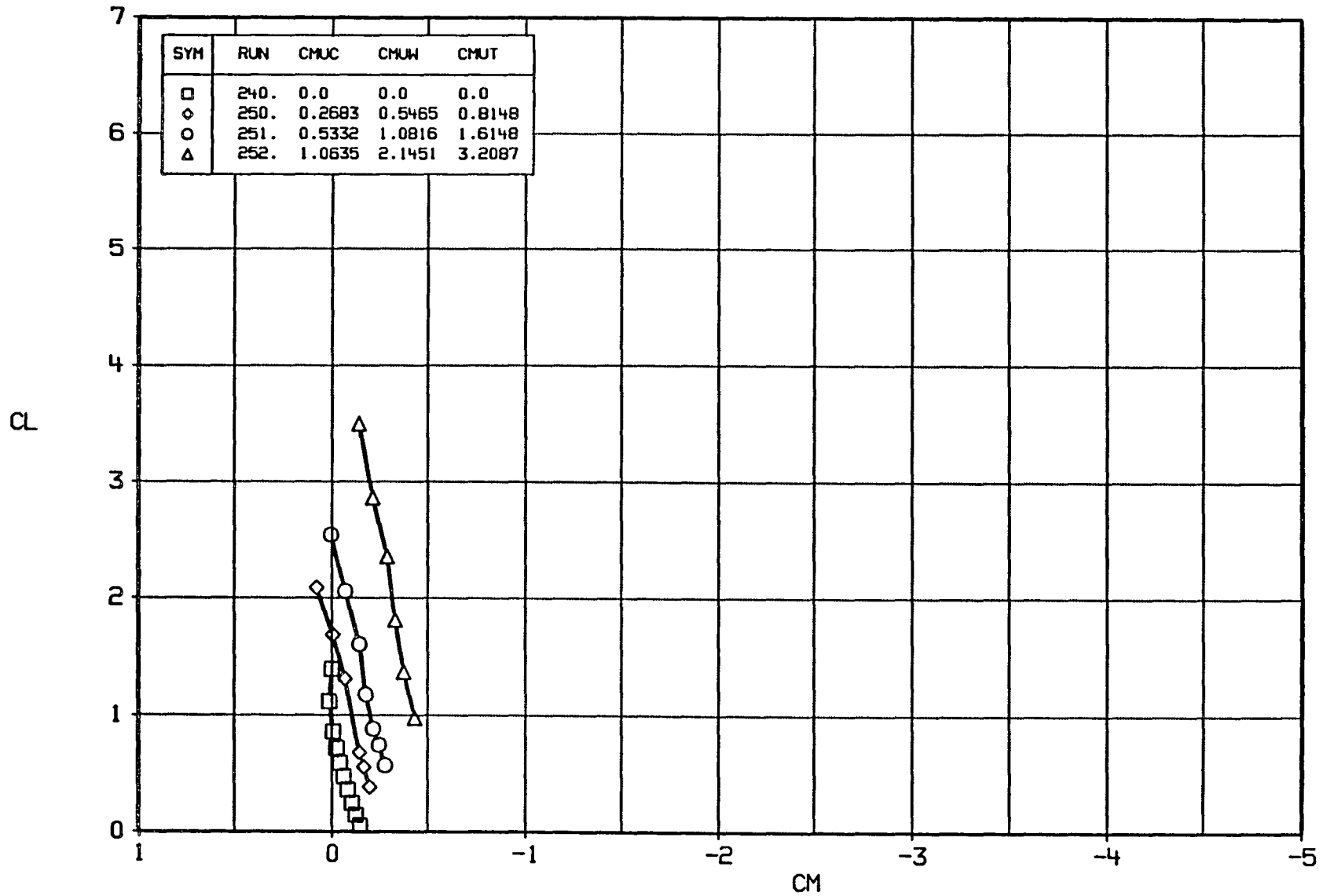


FIGURE A16c BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V,  $(BN/B)C=1$ ,  $(BN/B)W=0.5$ , DELF=0



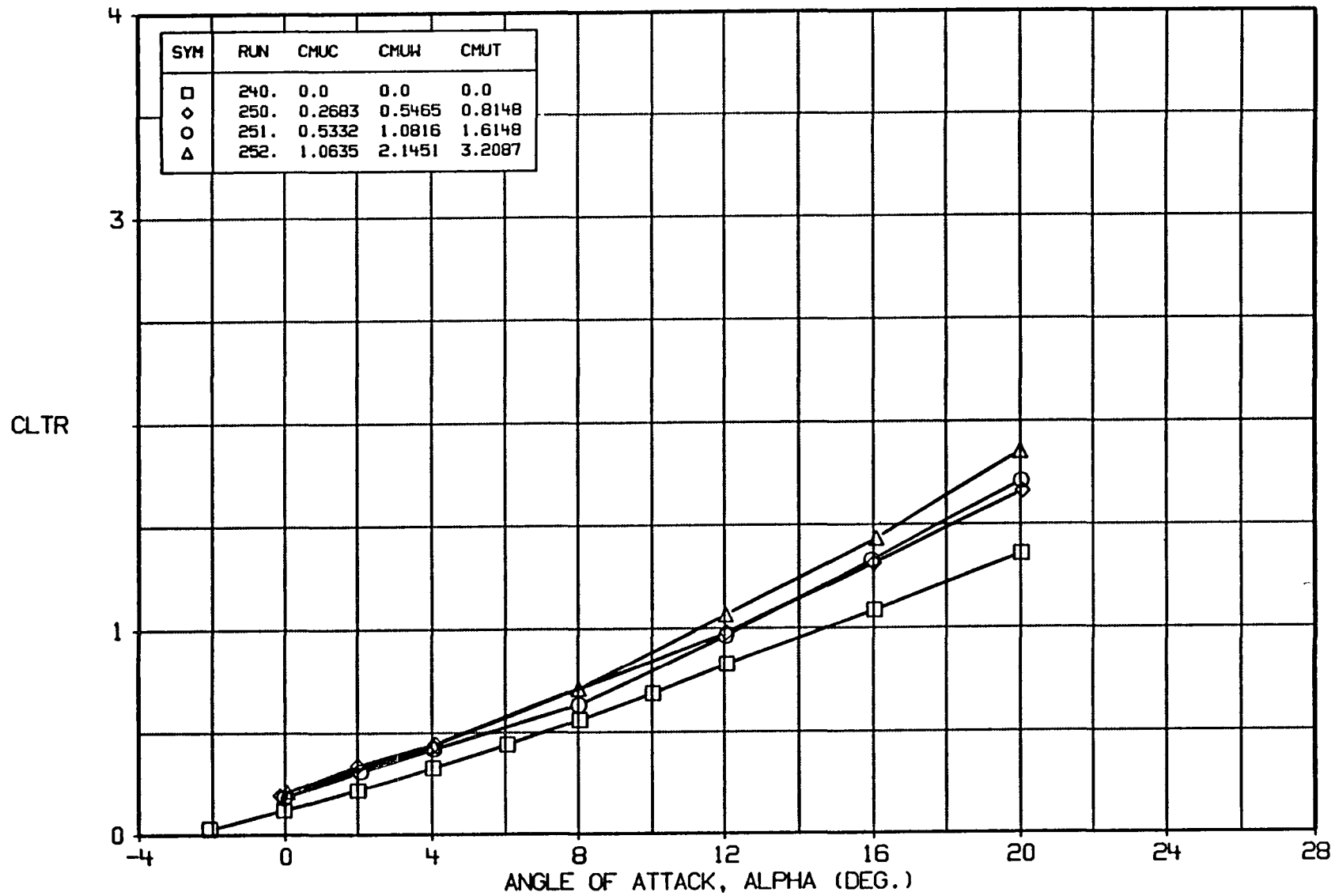


FIGURE A16d BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V,  $(BN/B)C=1$ ,  $(BN/B)W=0.5$ ,  $DEL F=0$

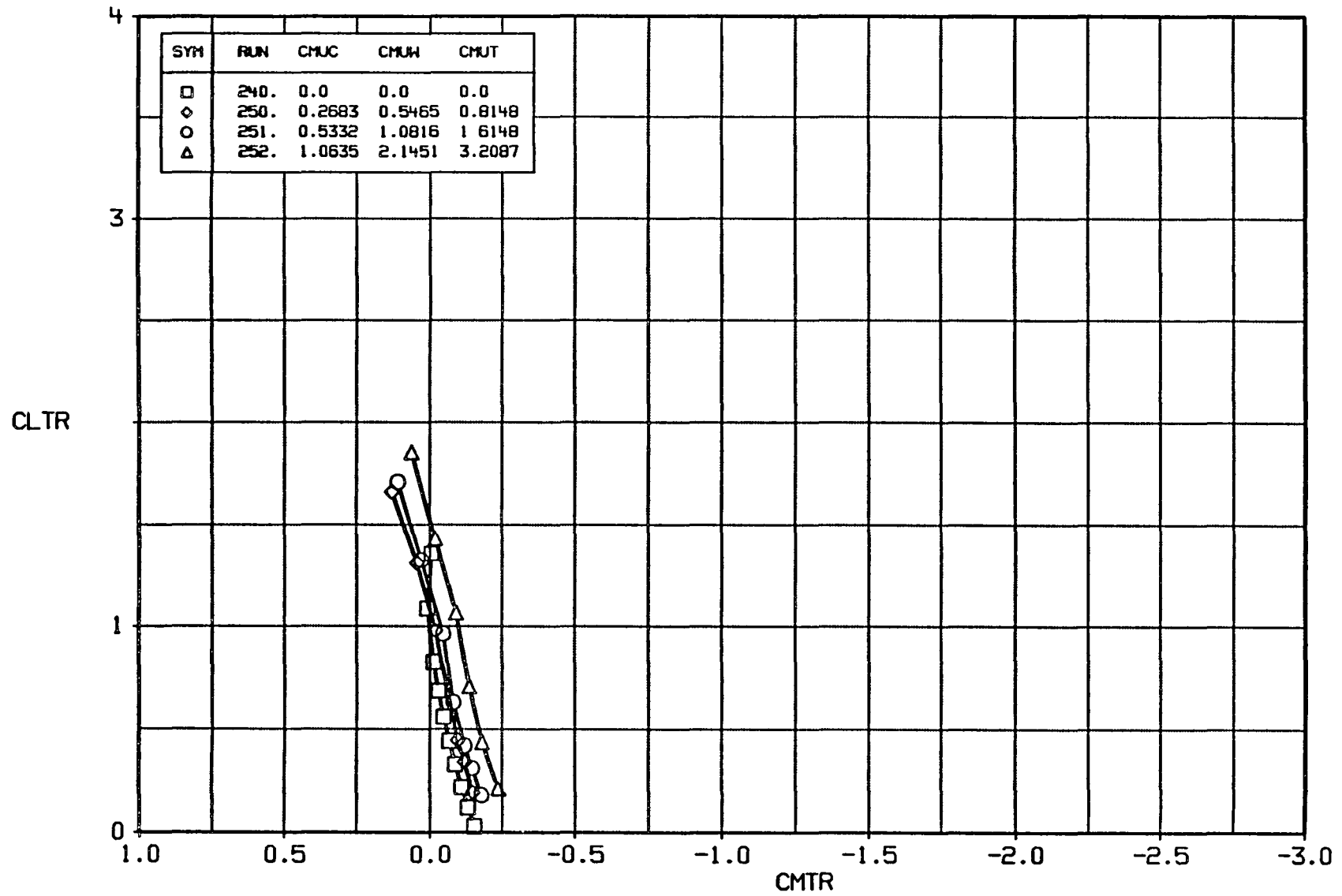


FIGURE A16e BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V,  $(BN/B)C=1$ ,  $(BN/B)W=0.5$ , DELF=0

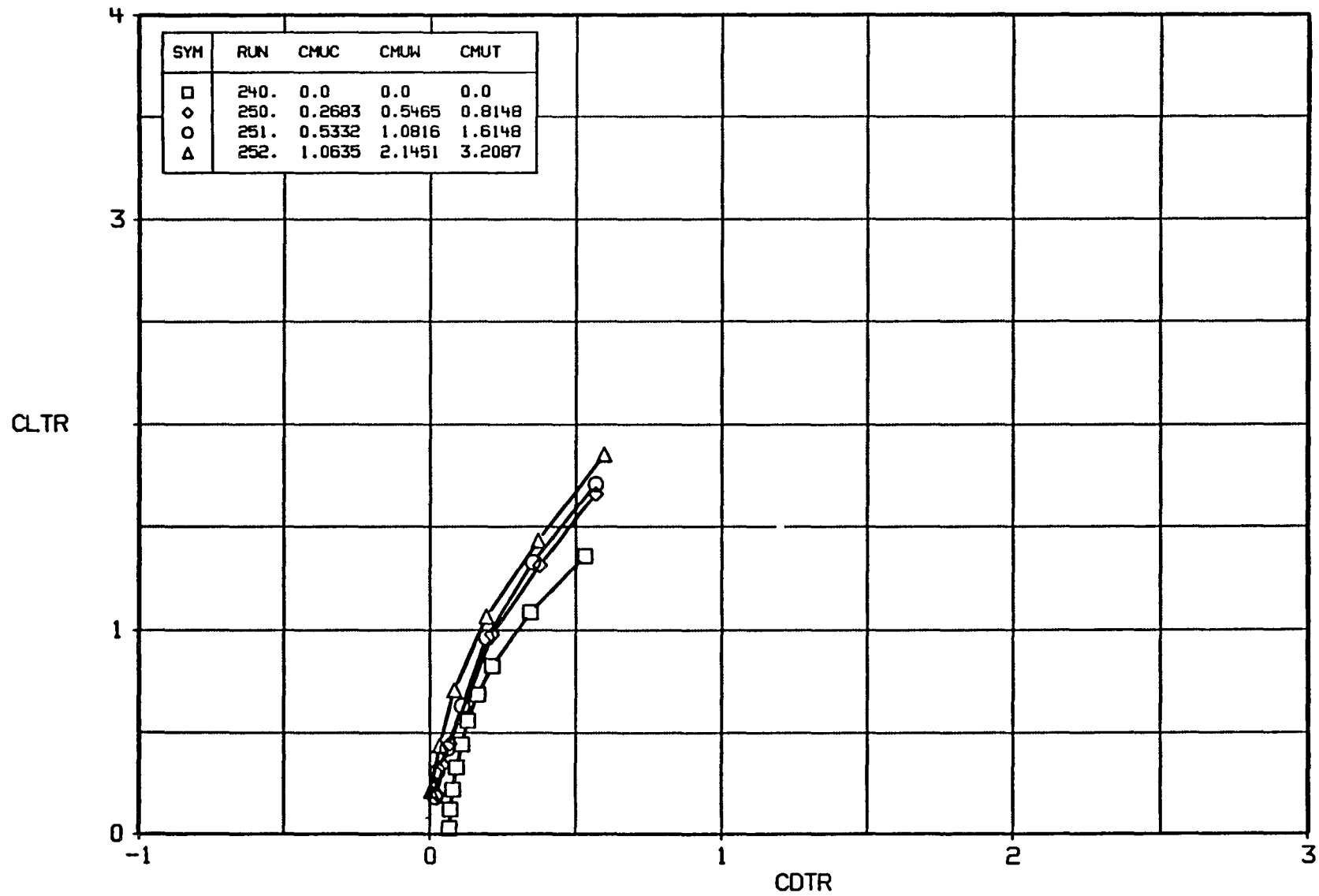


FIGURE A16f BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V,  $(BN/B)C=1$ ,  $(BN/B)W=0.5$ , DELF=0

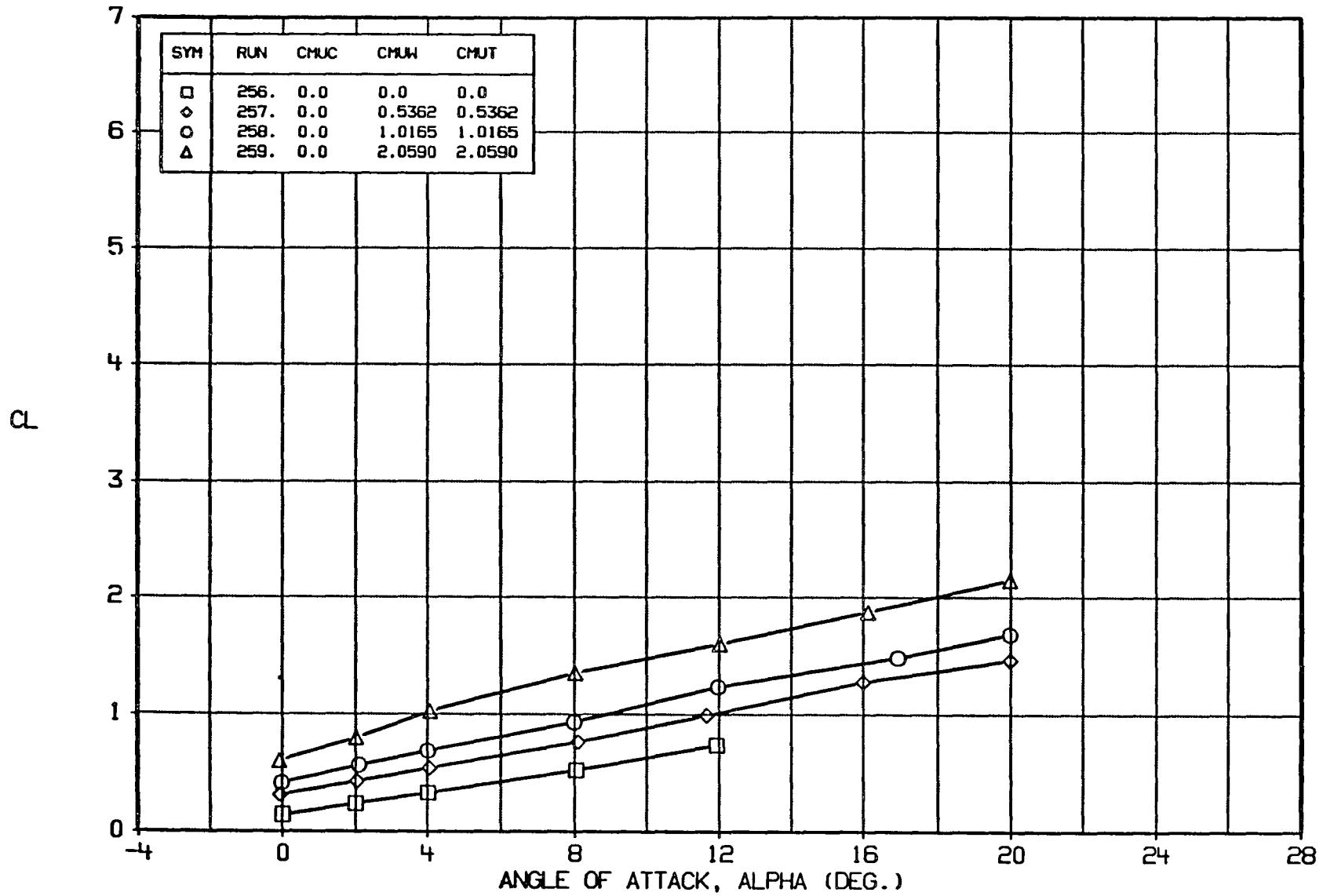


FIGURE A17a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V,  $BN/B=0.5$ ,  $DEL F=0$

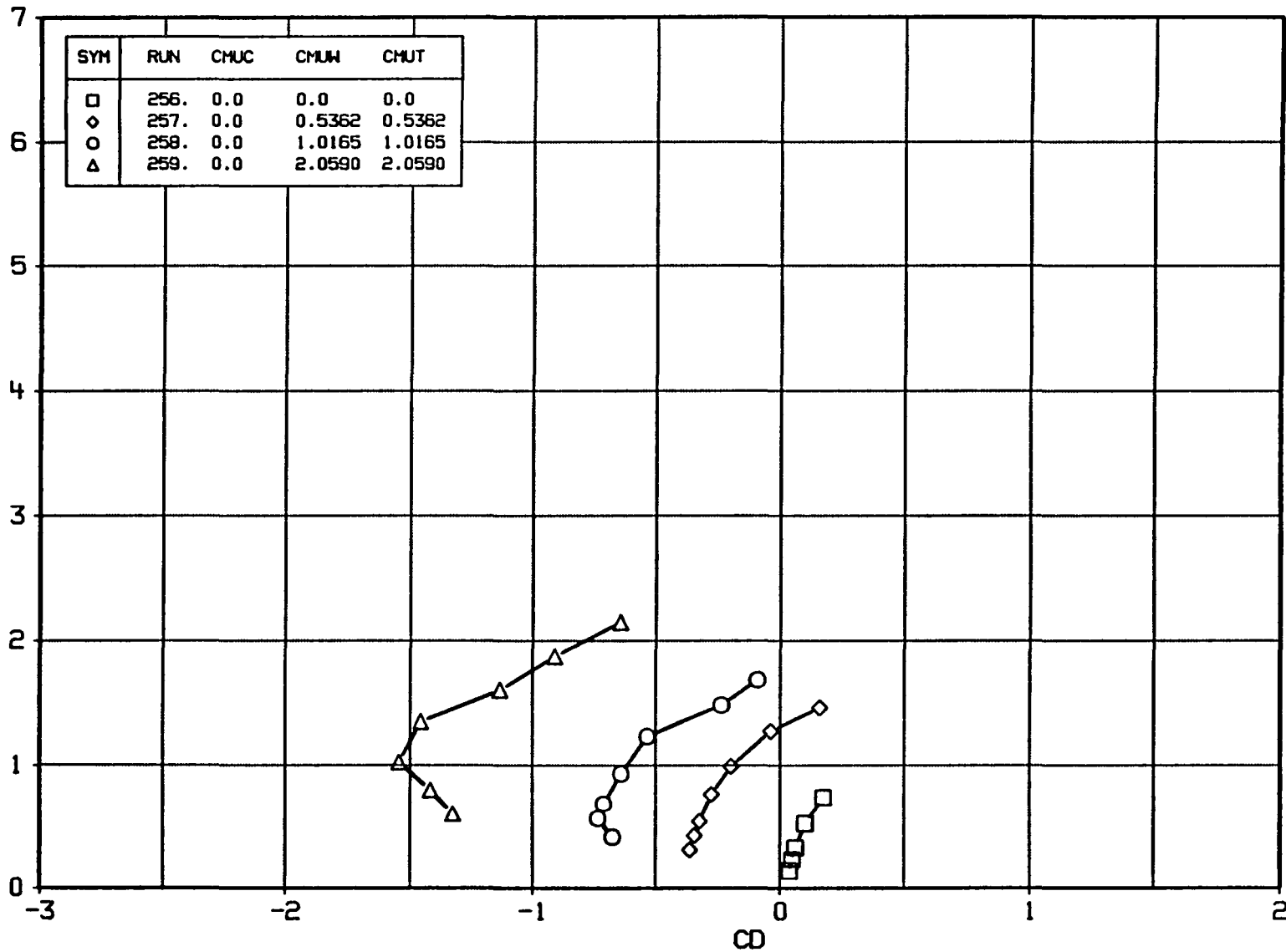


FIGURE A17b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5, DELF=0

A-108

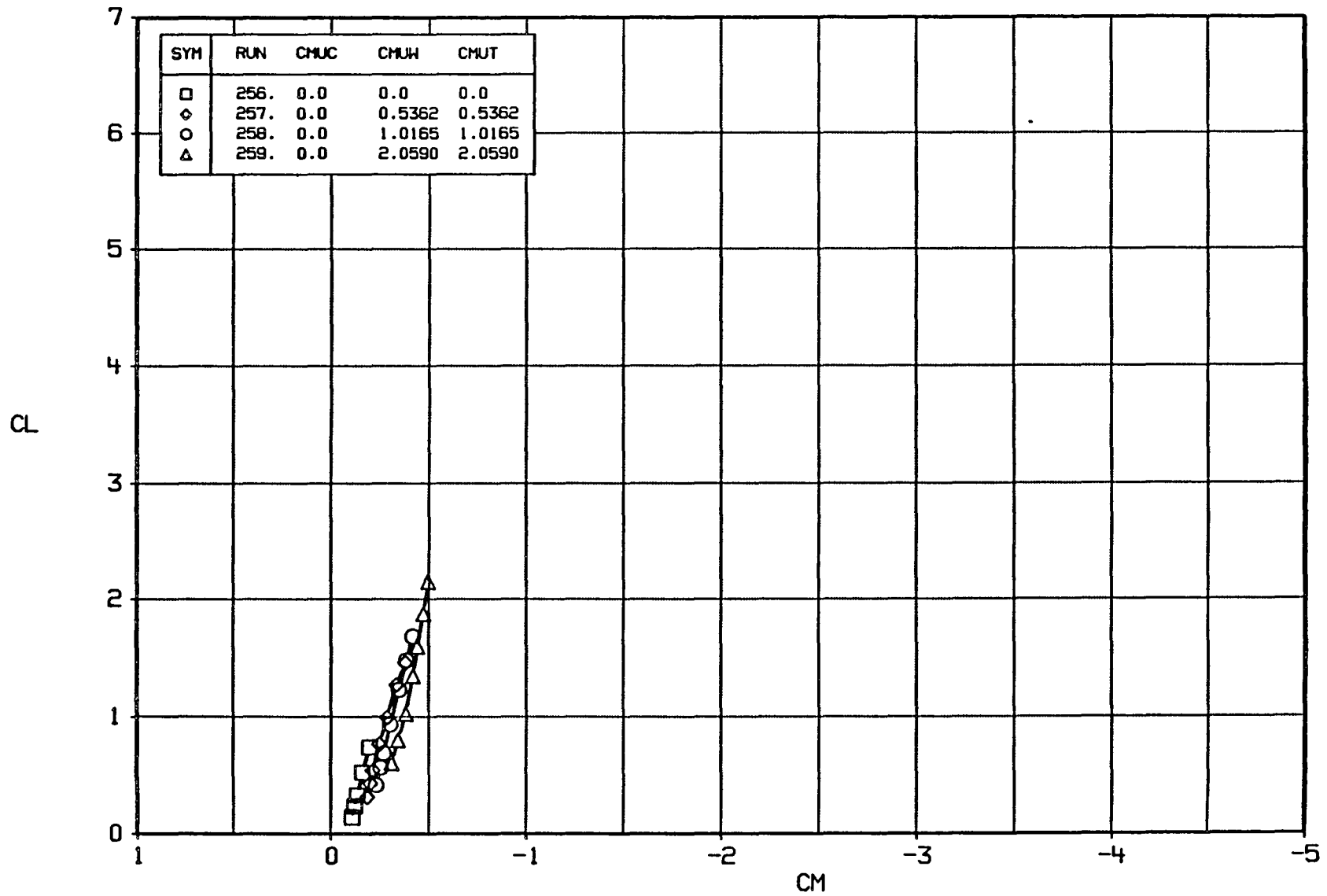


FIGURE A17c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V,  $BN/B=0.5$ ,  $DEL F=0$

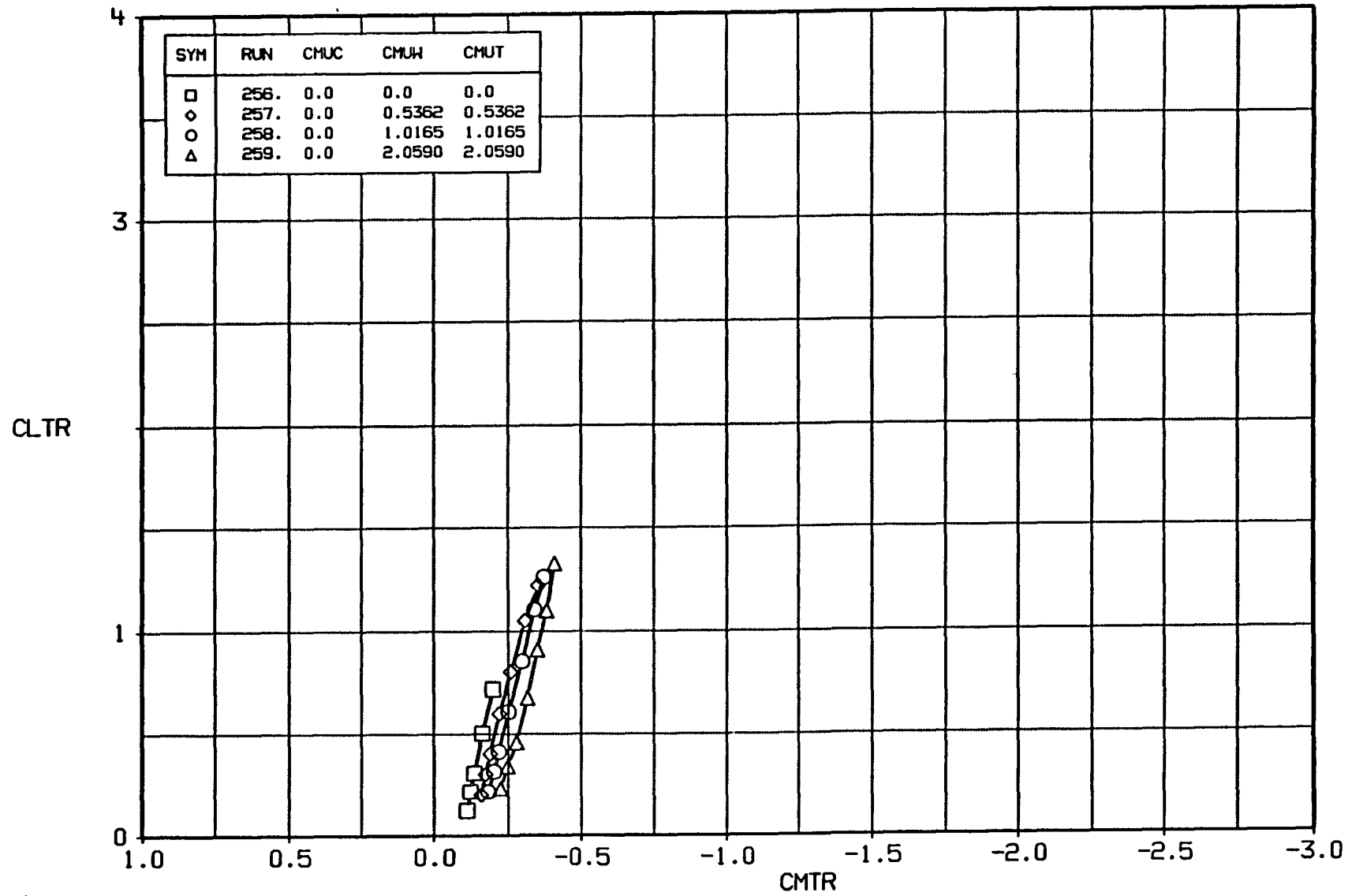


FIGURE A17d BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V,  $BN/B=0.5$ , DELF=0

A-110

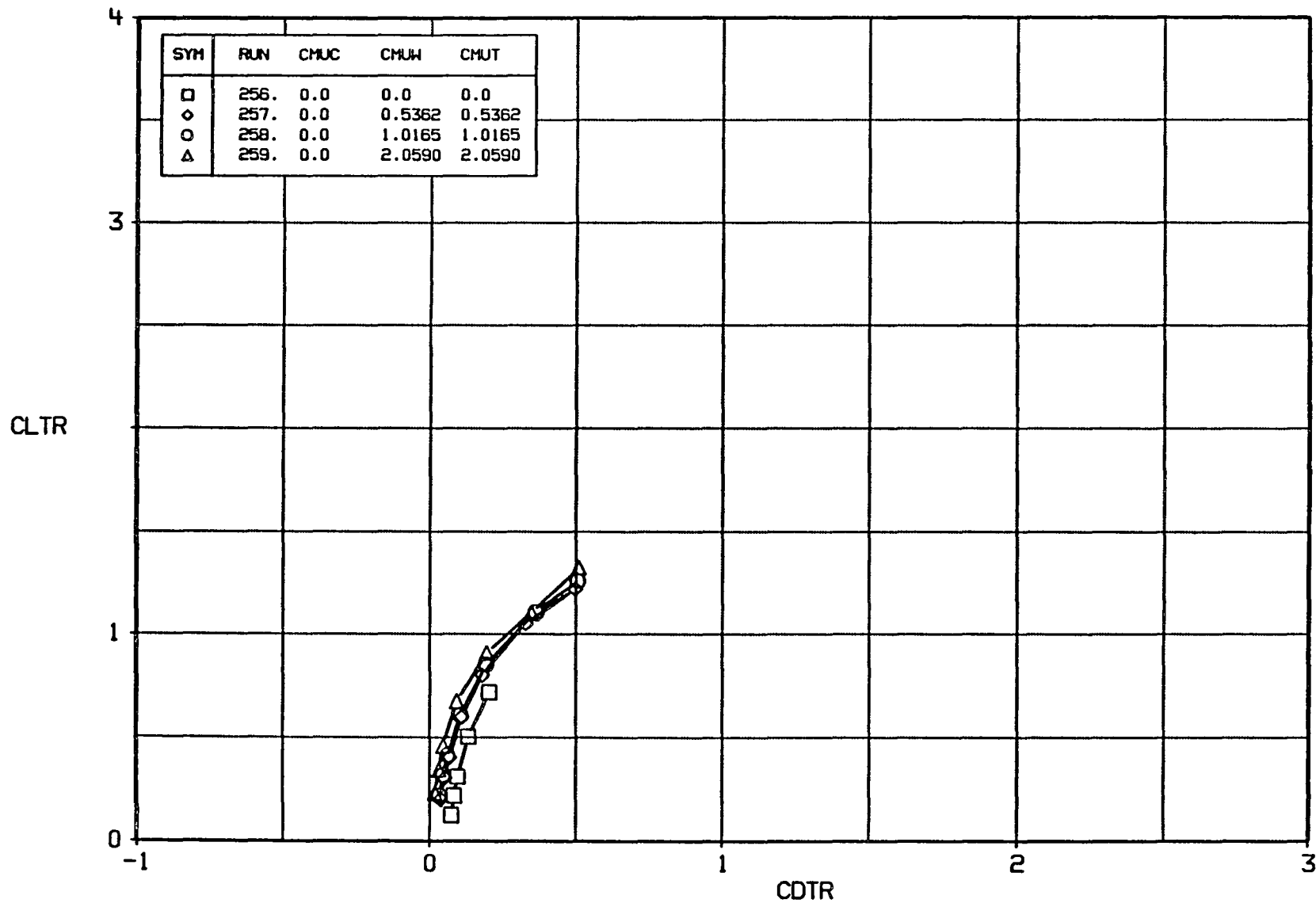


FIGURE A17e BASIC DATA EFFECT OF CMU  
CONFIGURATION BWSV,  $BN/B=0.5$ , DELF=0



A-111

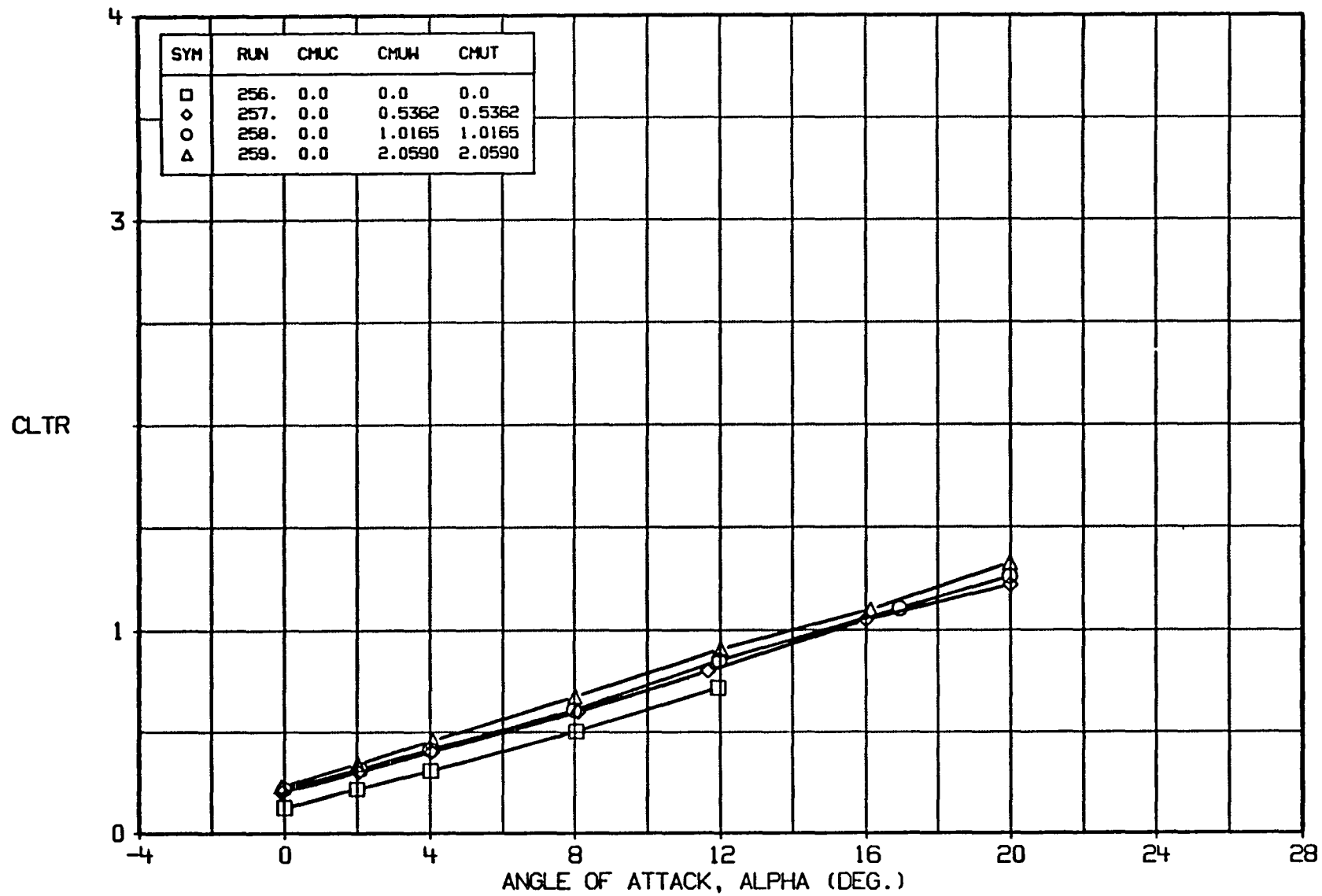


FIGURE A17f BASIC DATA EFFECT OF CMU CONFIGURATION BWSV,  $B_N/B=0.5$ ,  $\Delta E_L=0$

A-112

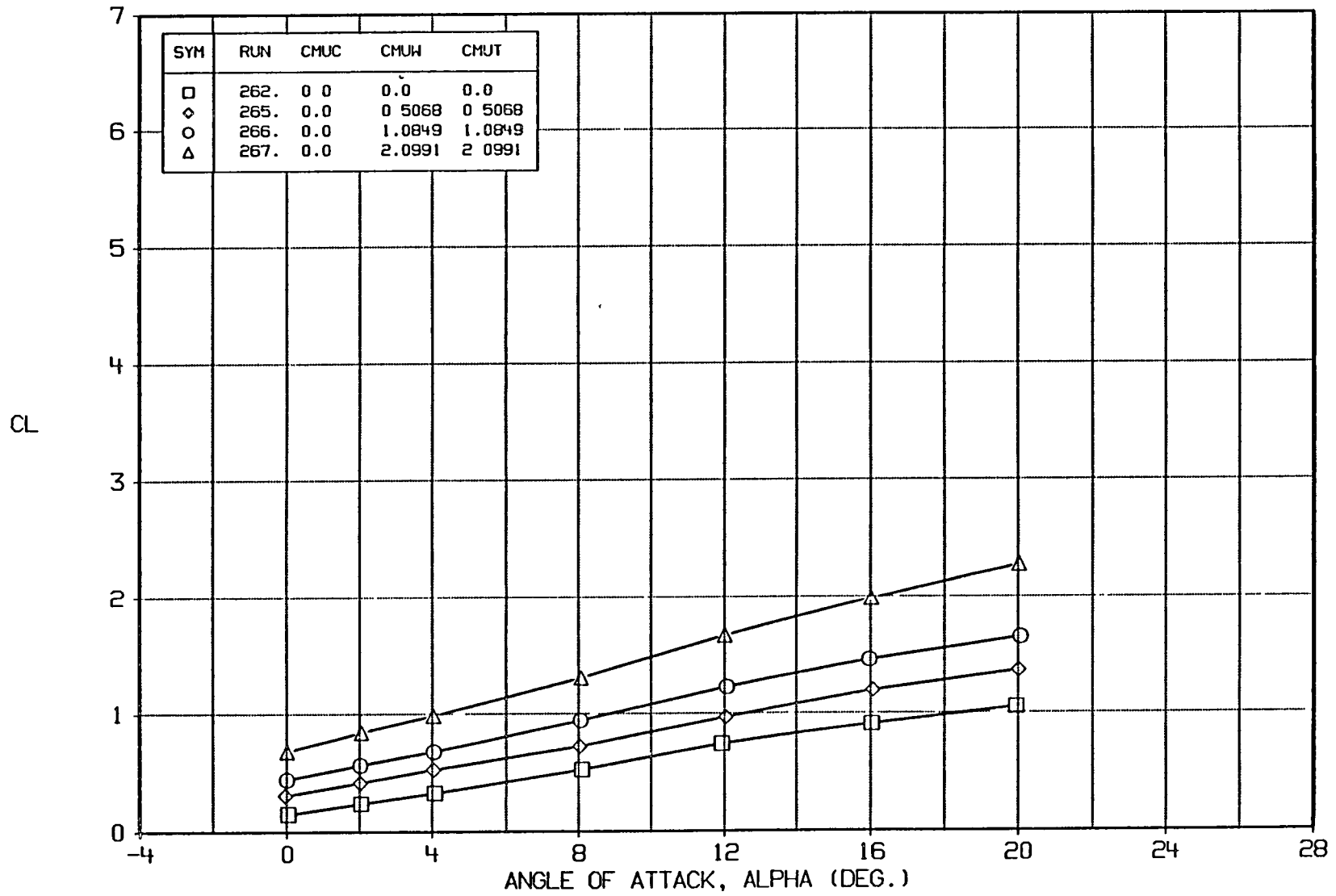


FIGURE A18a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.25, DELF=0

A-113

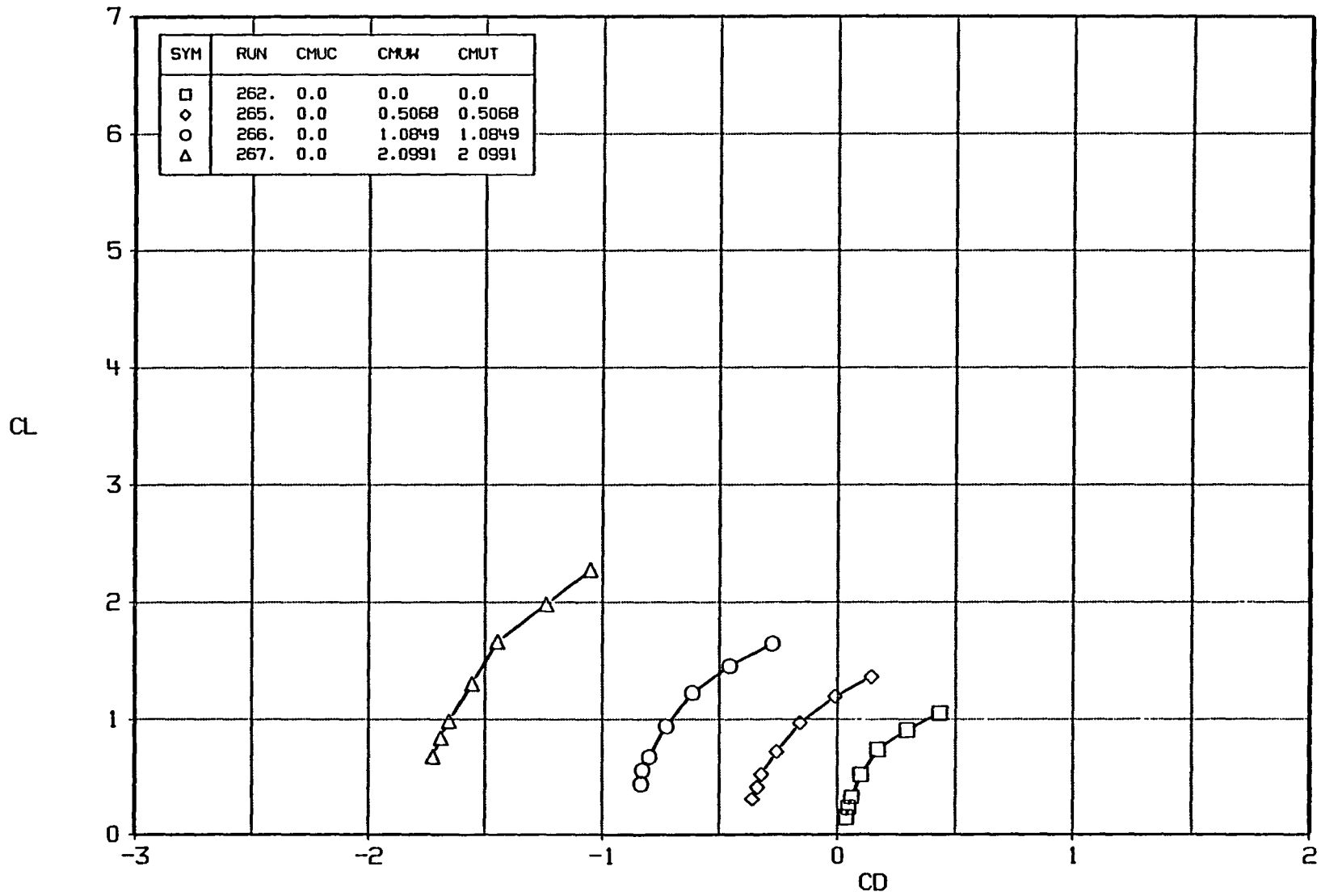


FIGURE A18b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.25, DELF=0

A-114

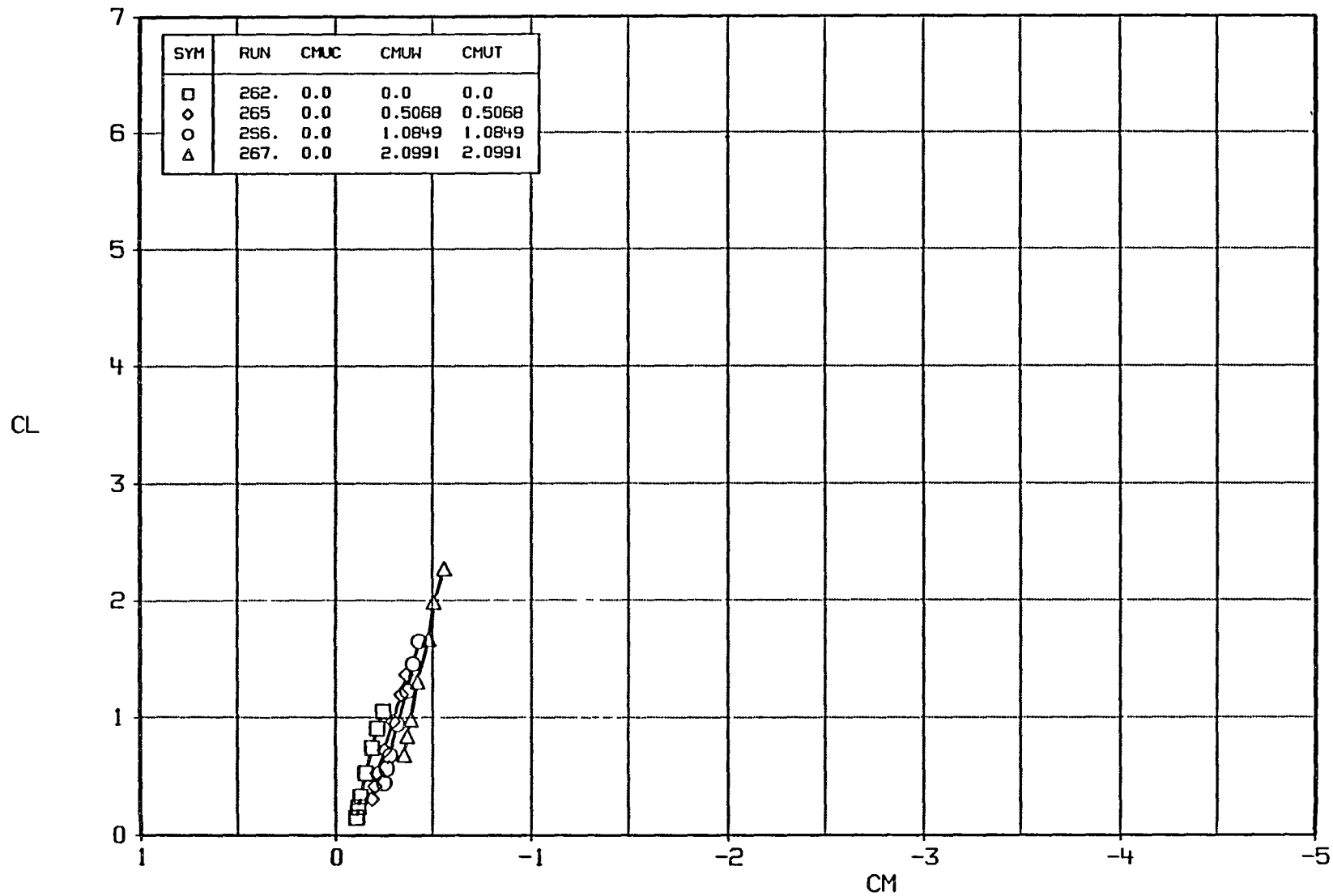


FIGURE A18c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.25, DELF=0

A-115

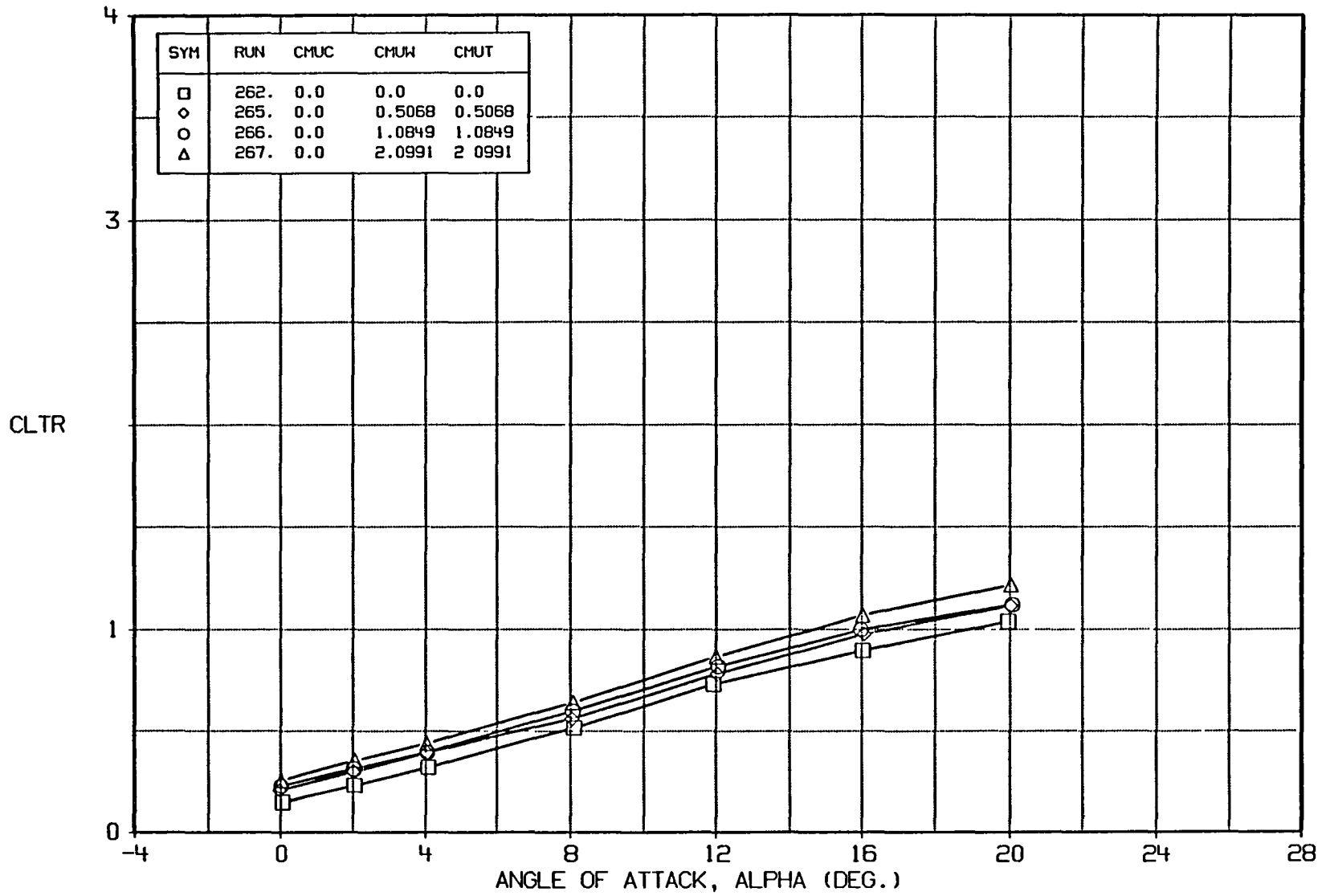


FIGURE A18d BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.25, DELF=0

A-116

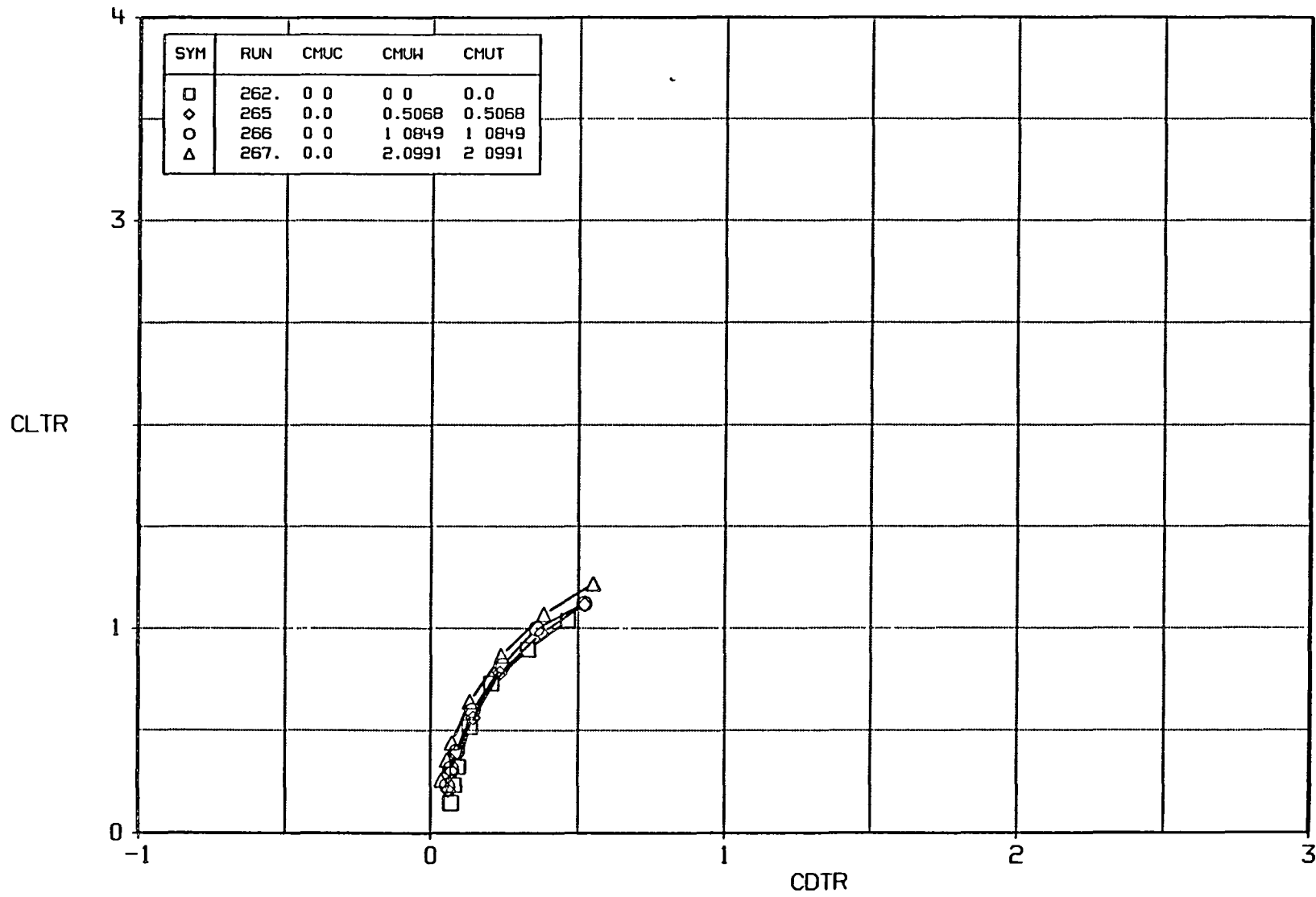


FIGURE A18e BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.25, DELF=0

A-117

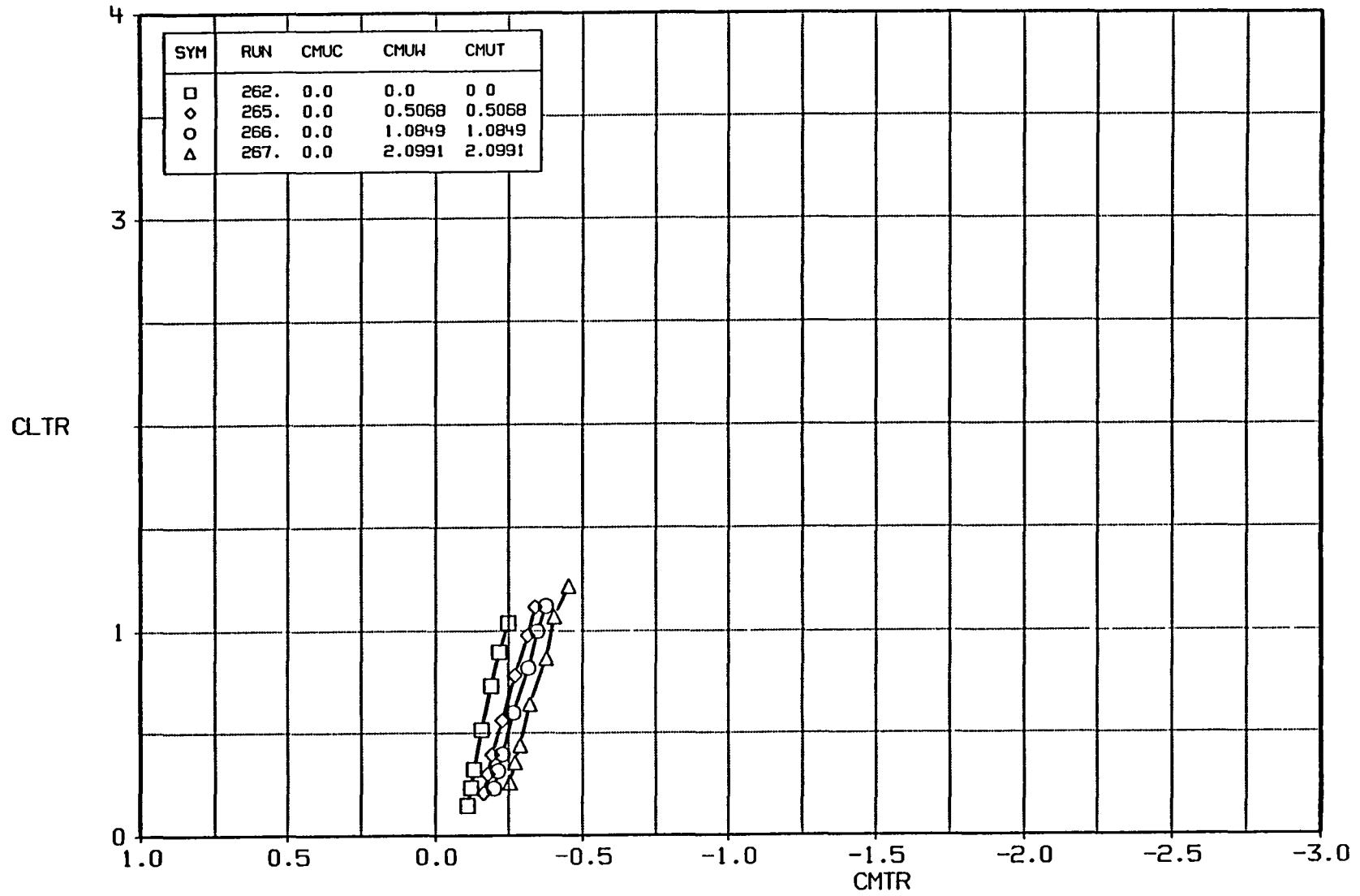


FIGURE A18f BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.25, DELF=0

A-118

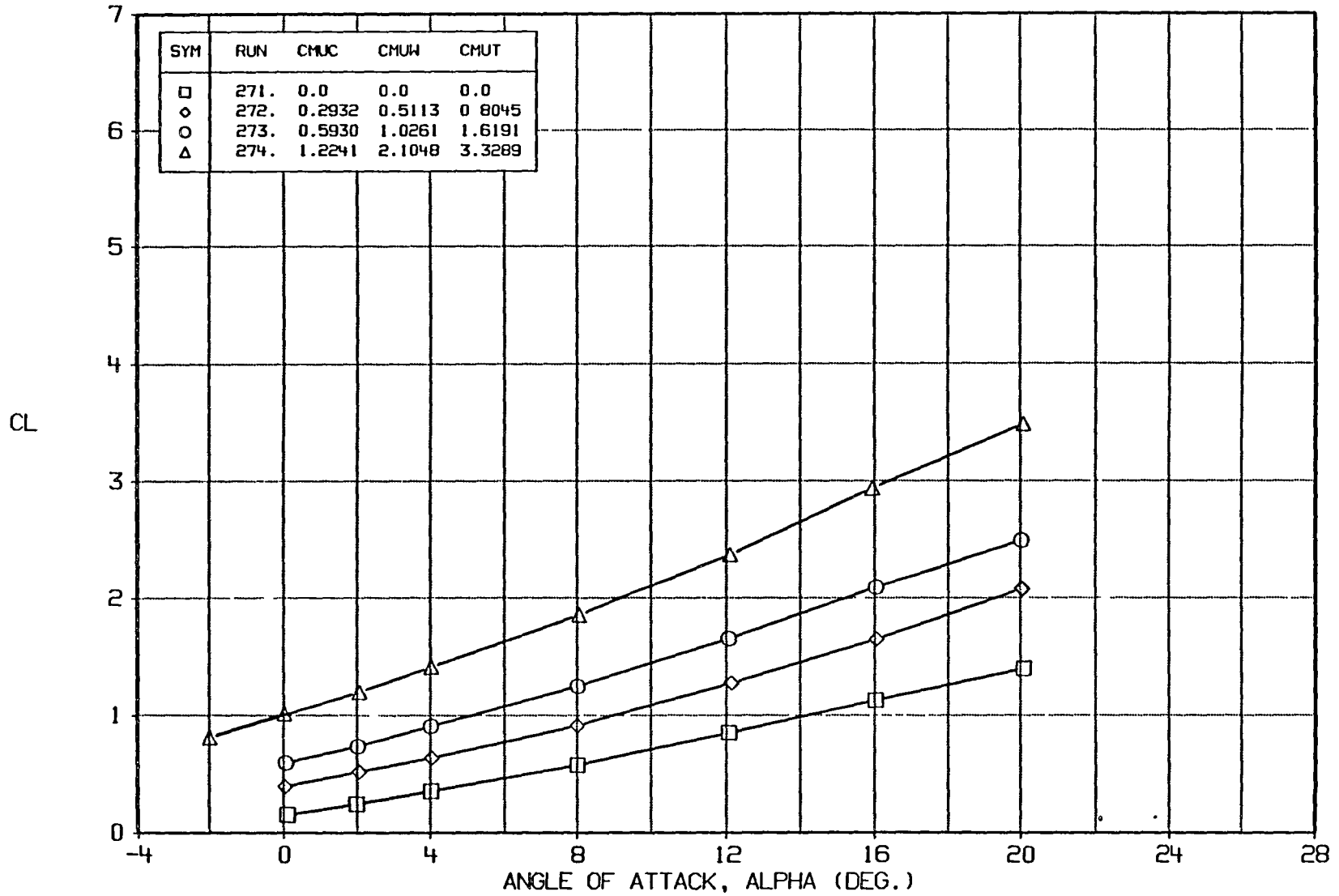


FIGURE A19a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.25. DELF=0



A-119

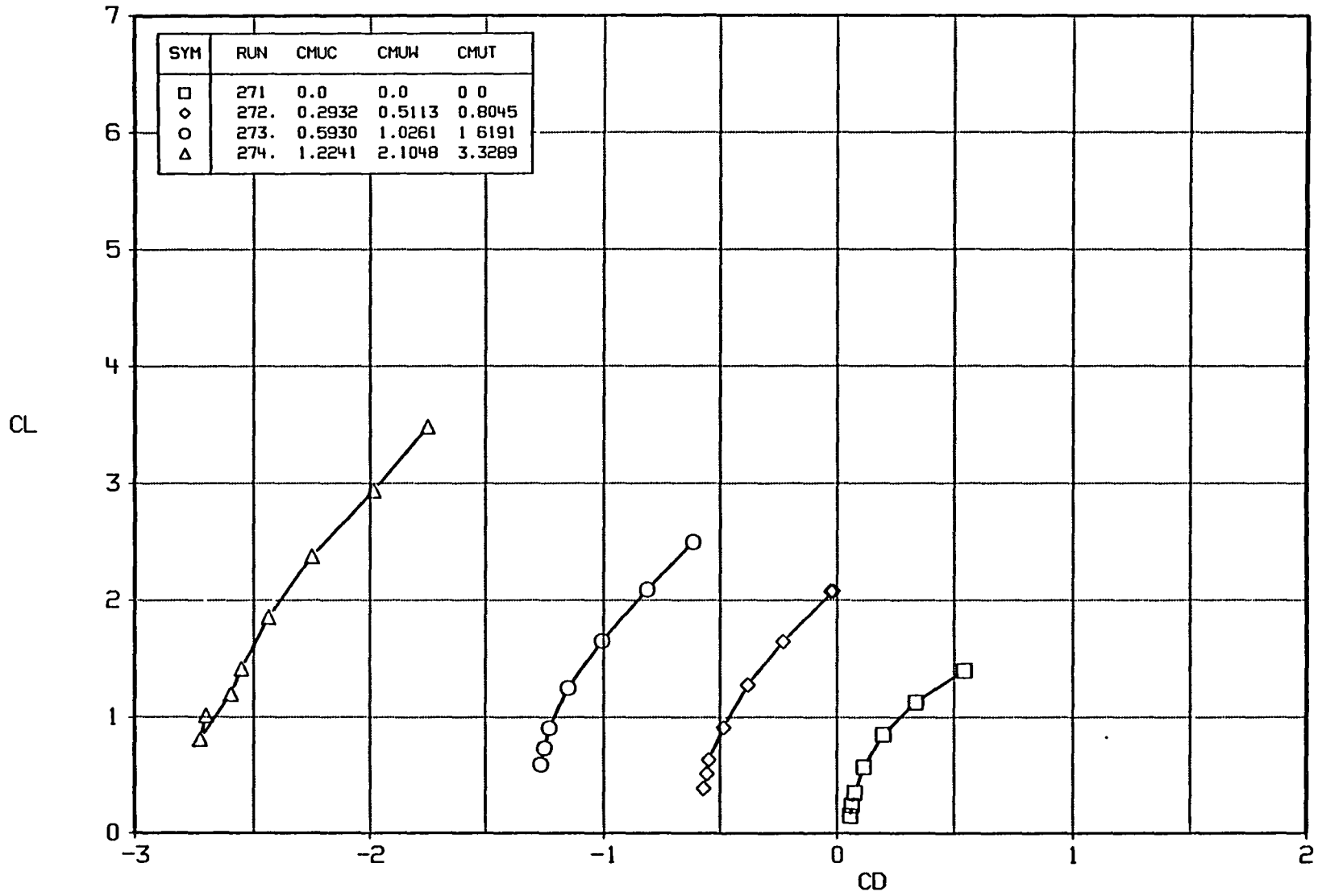


FIGURE A19b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.25, DELF=0

A-120

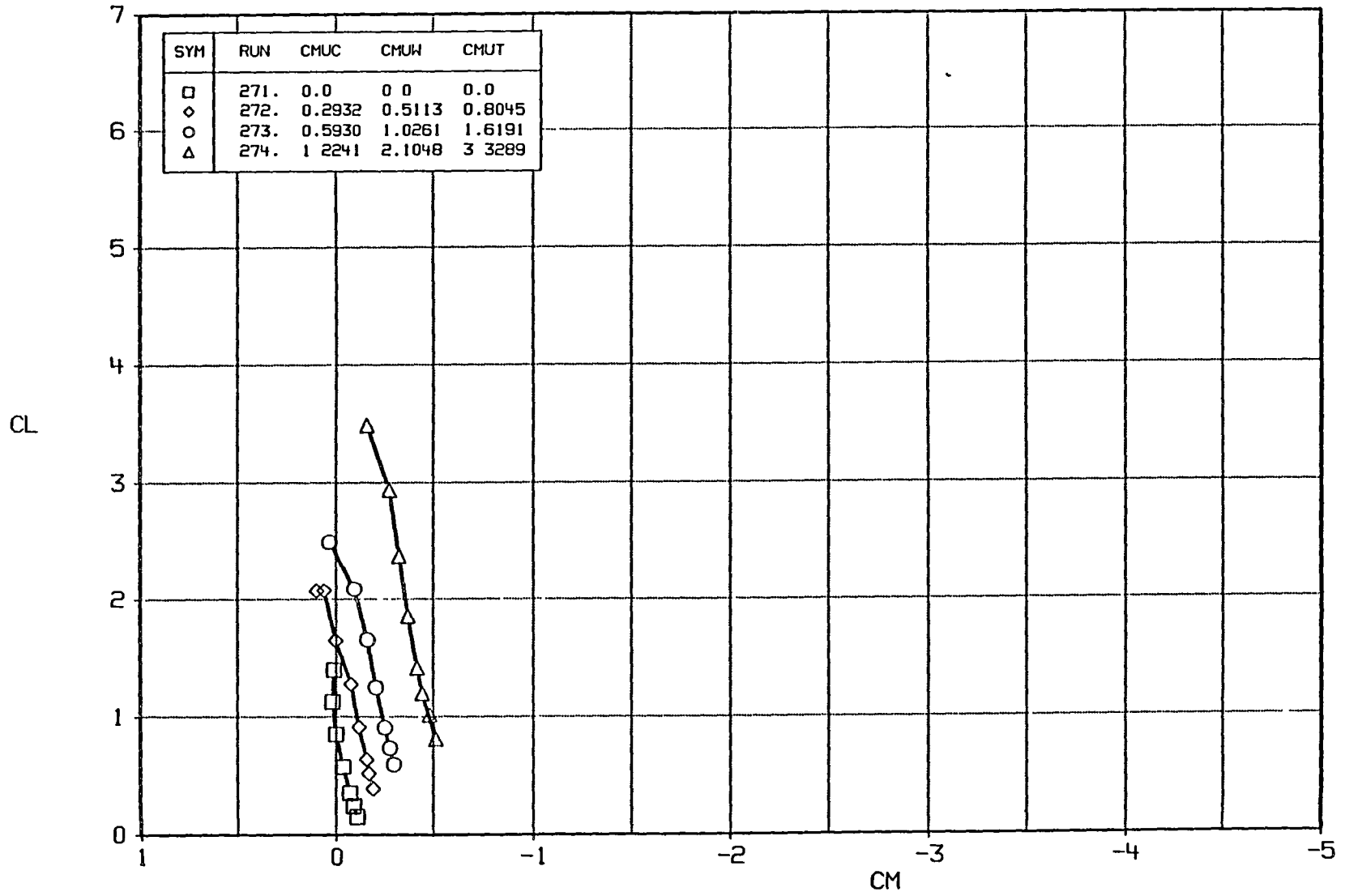


FIGURE A19c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.25, DELF=0

A-121

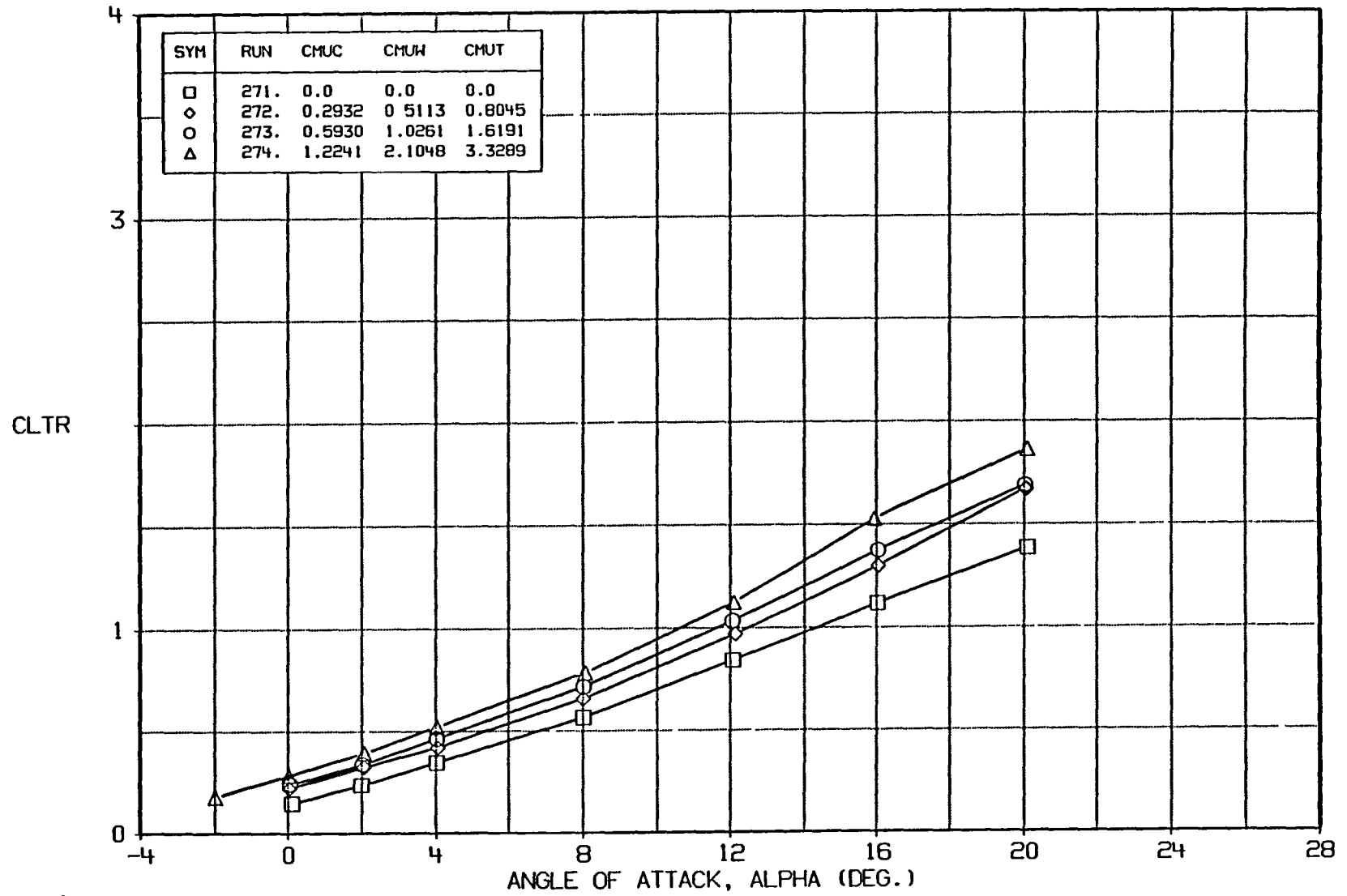


FIGURE A19d BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.25, DELF=0

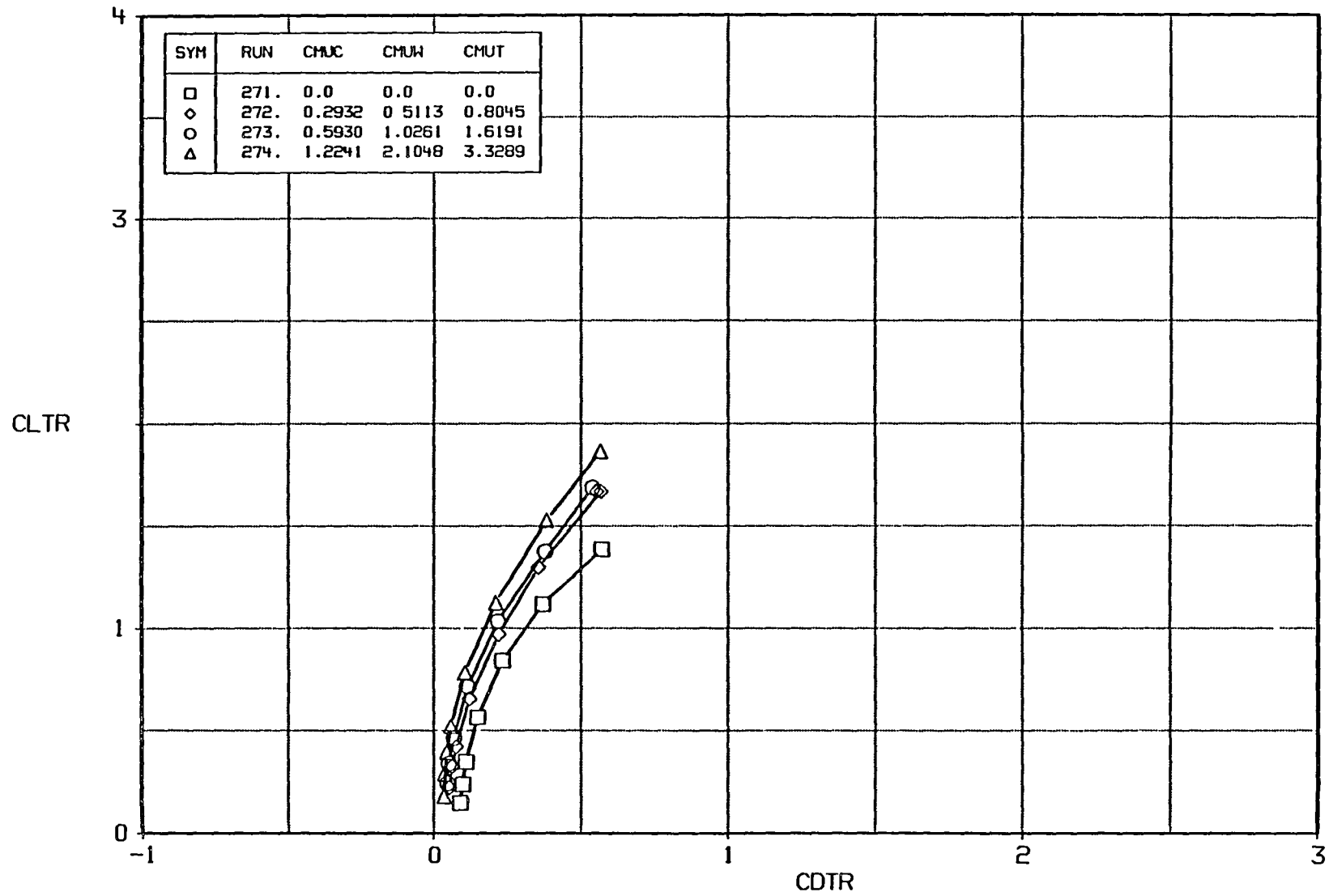


FIGURE A19e BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.25, DELF=0

A-123

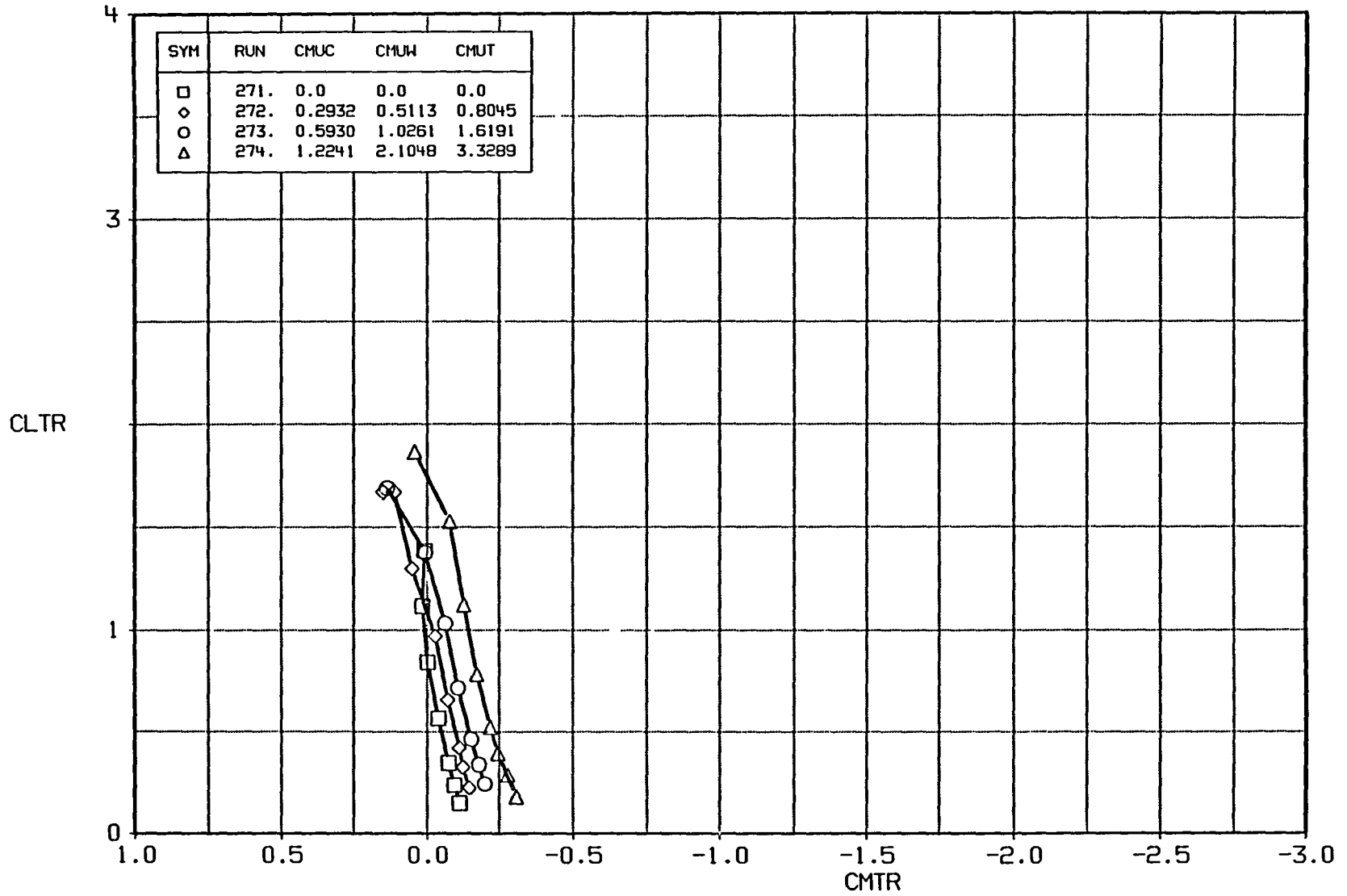


FIGURE A19f BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.25, DELF=0

A-124

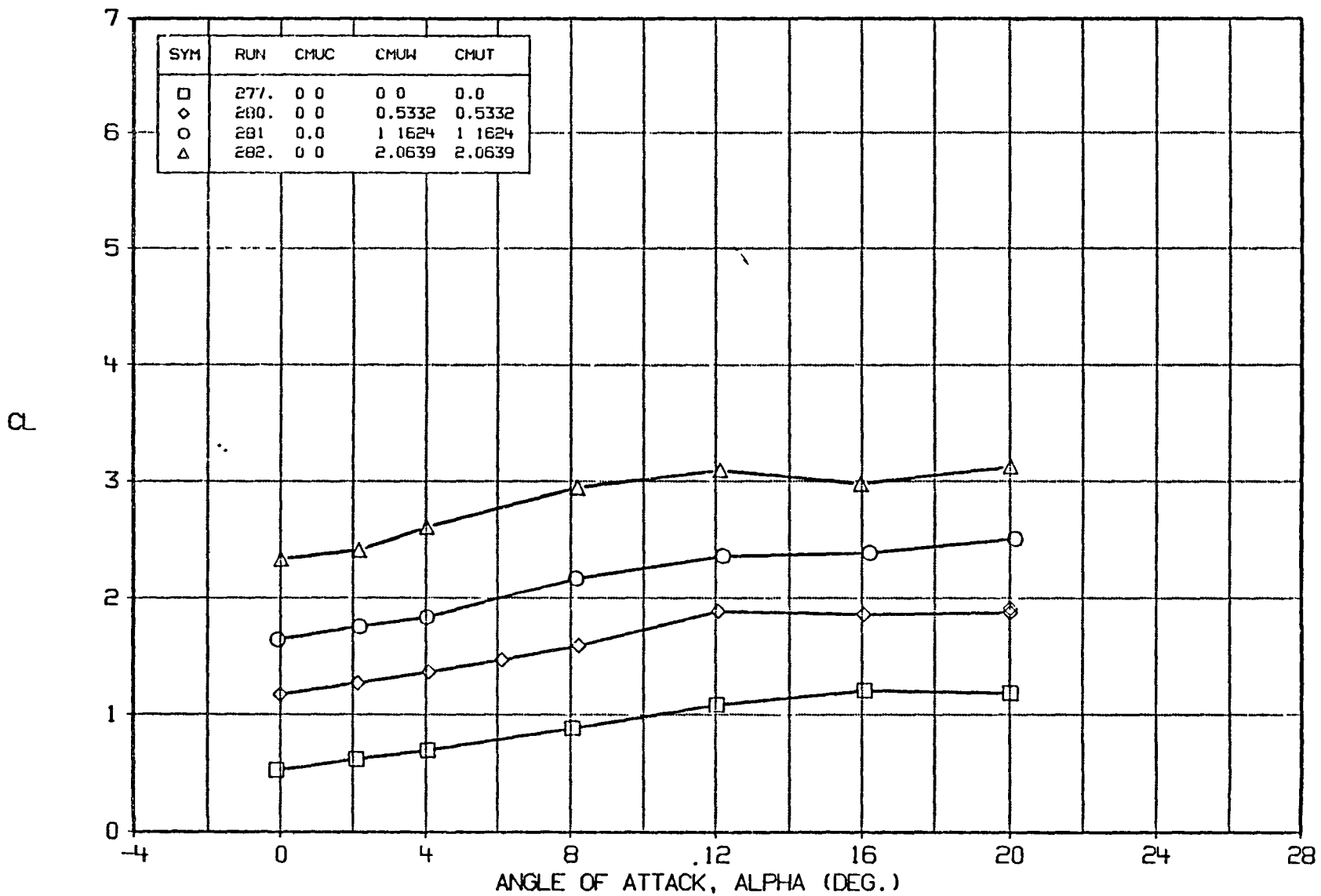


FIGURE A20a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25

A-125

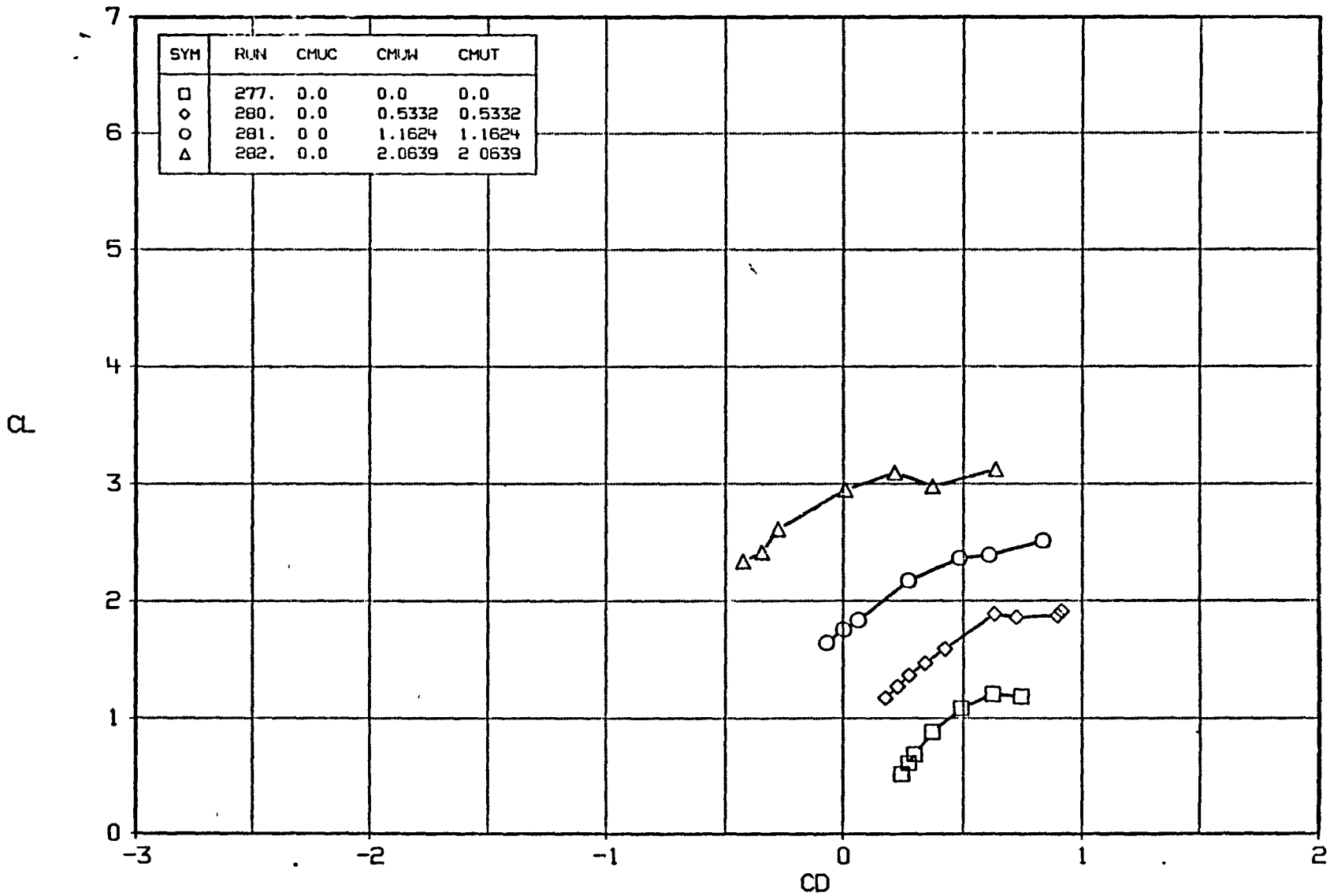


FIGURE A20b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25

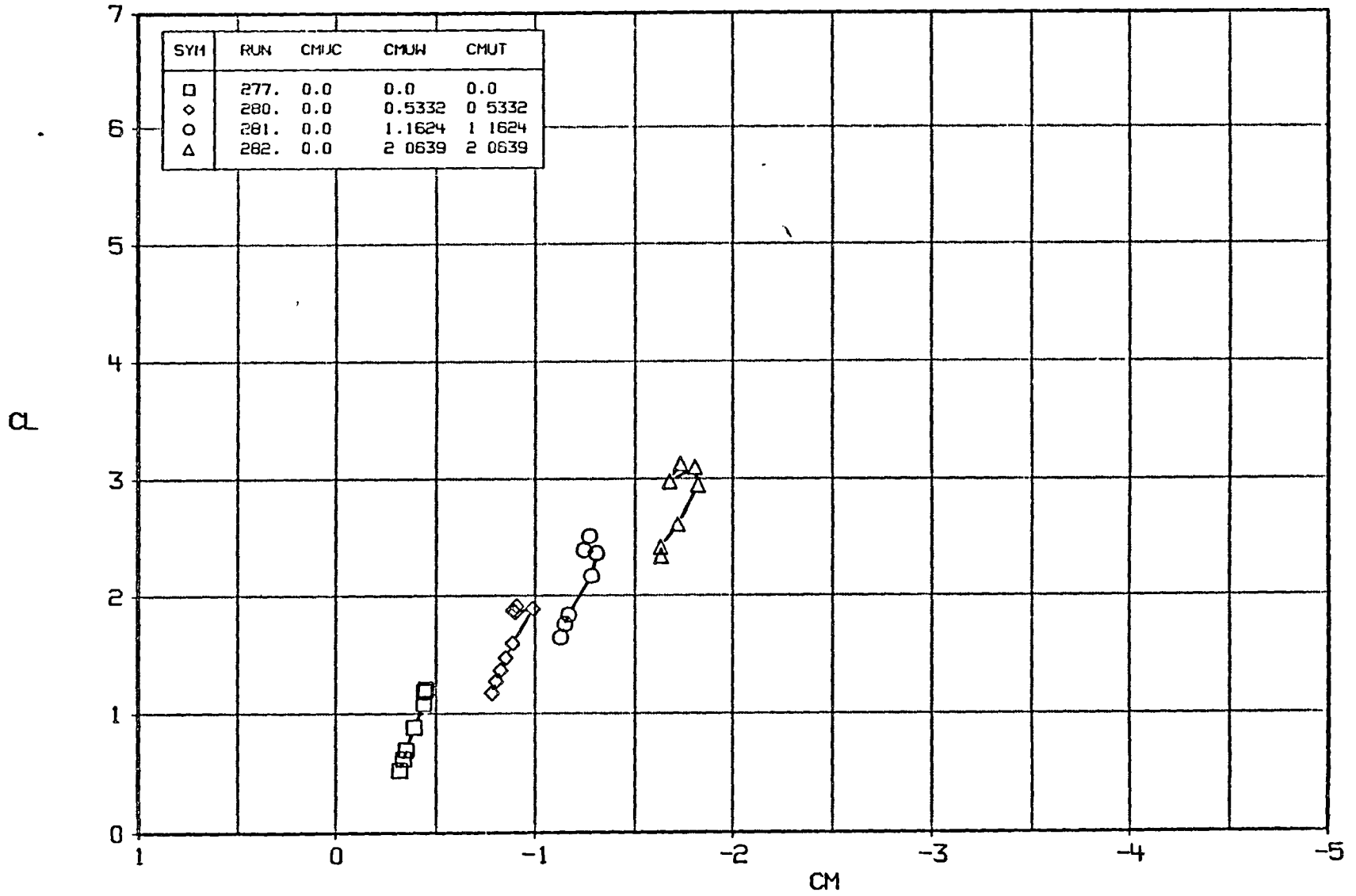


FIGURE A20c BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V, DELF=45, BN/B=.25



A-127

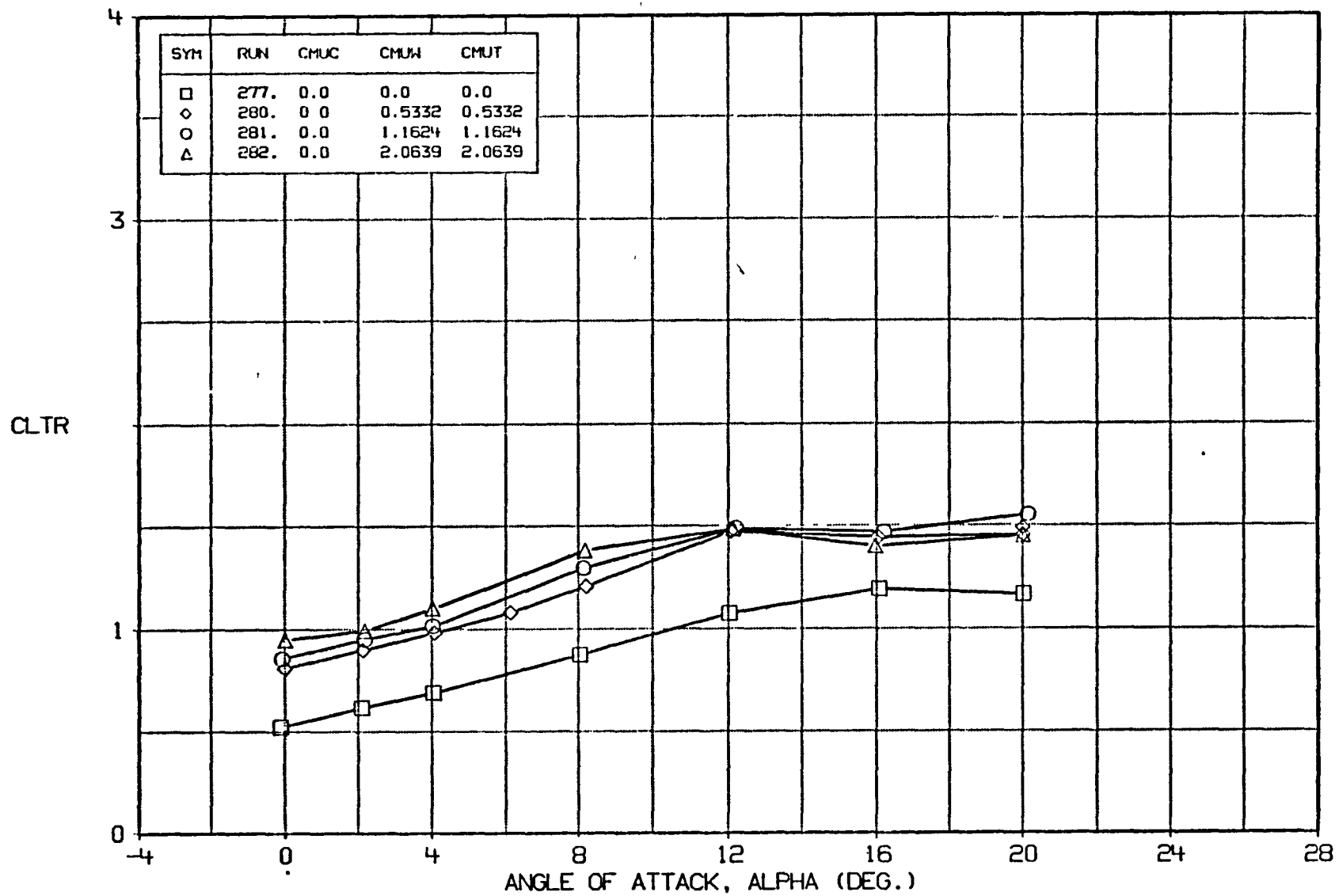


FIGURE A20d BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25

A-128

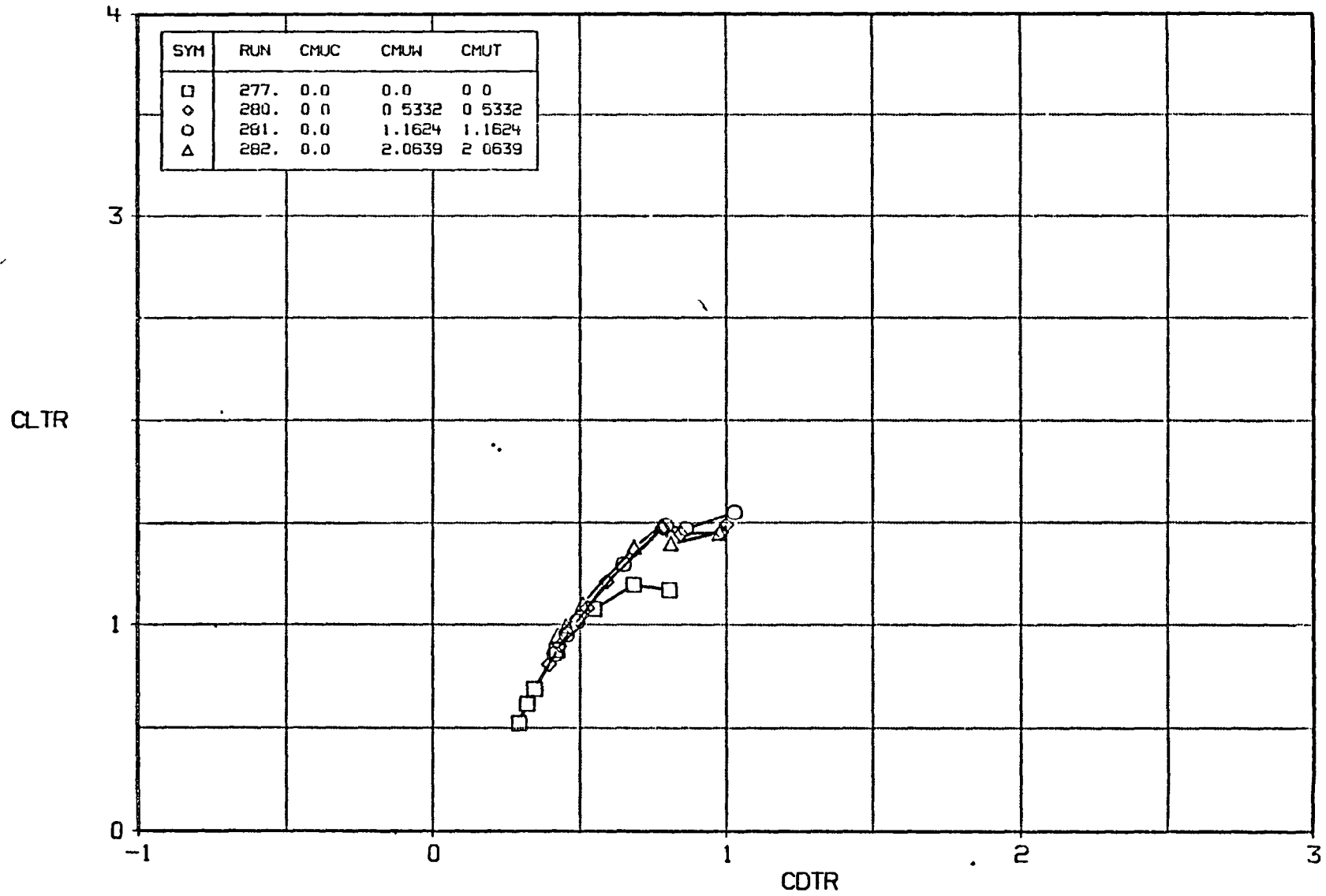


FIGURE A20e BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25

A-129

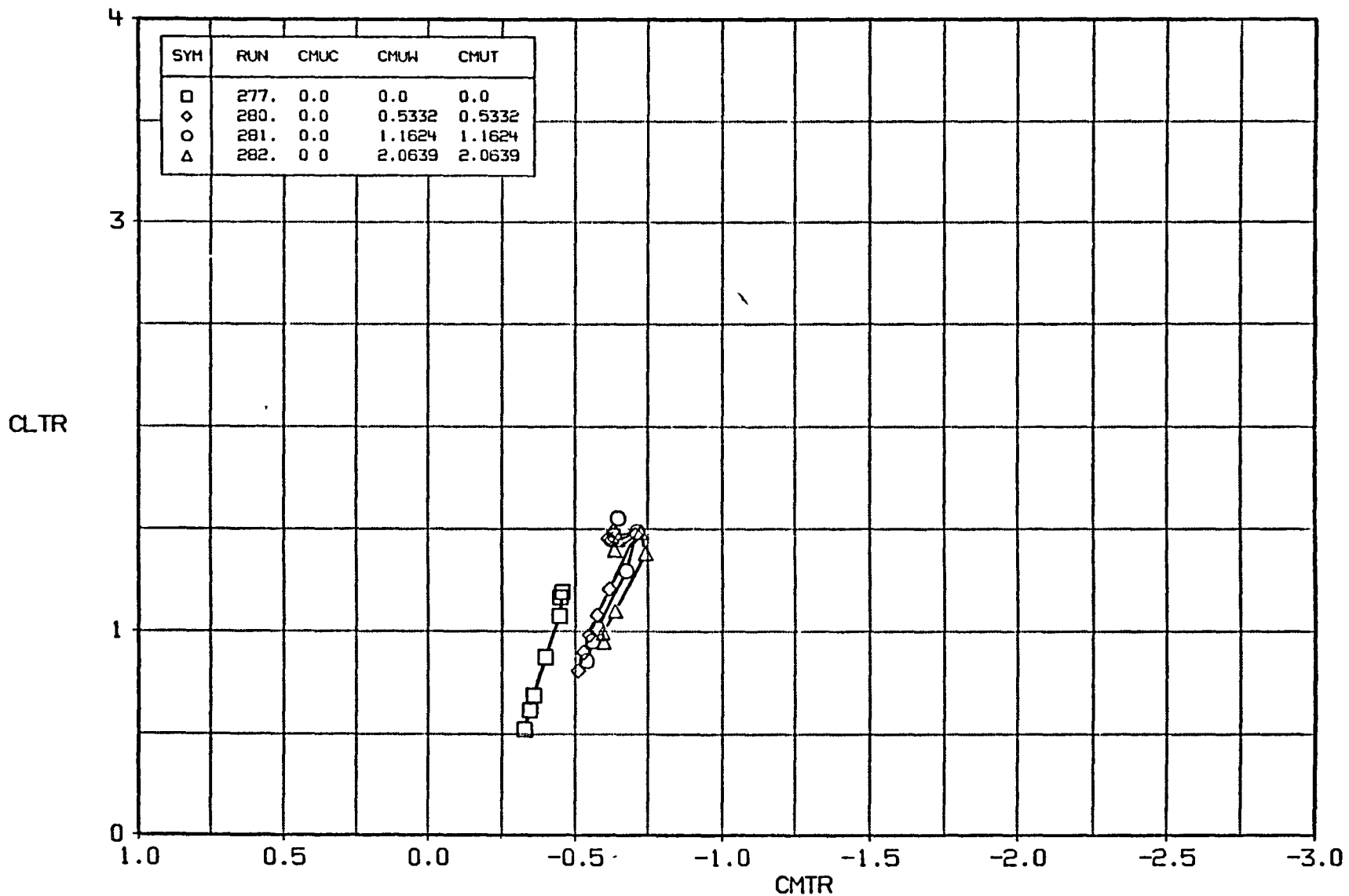


FIGURE A20f BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25

A-130

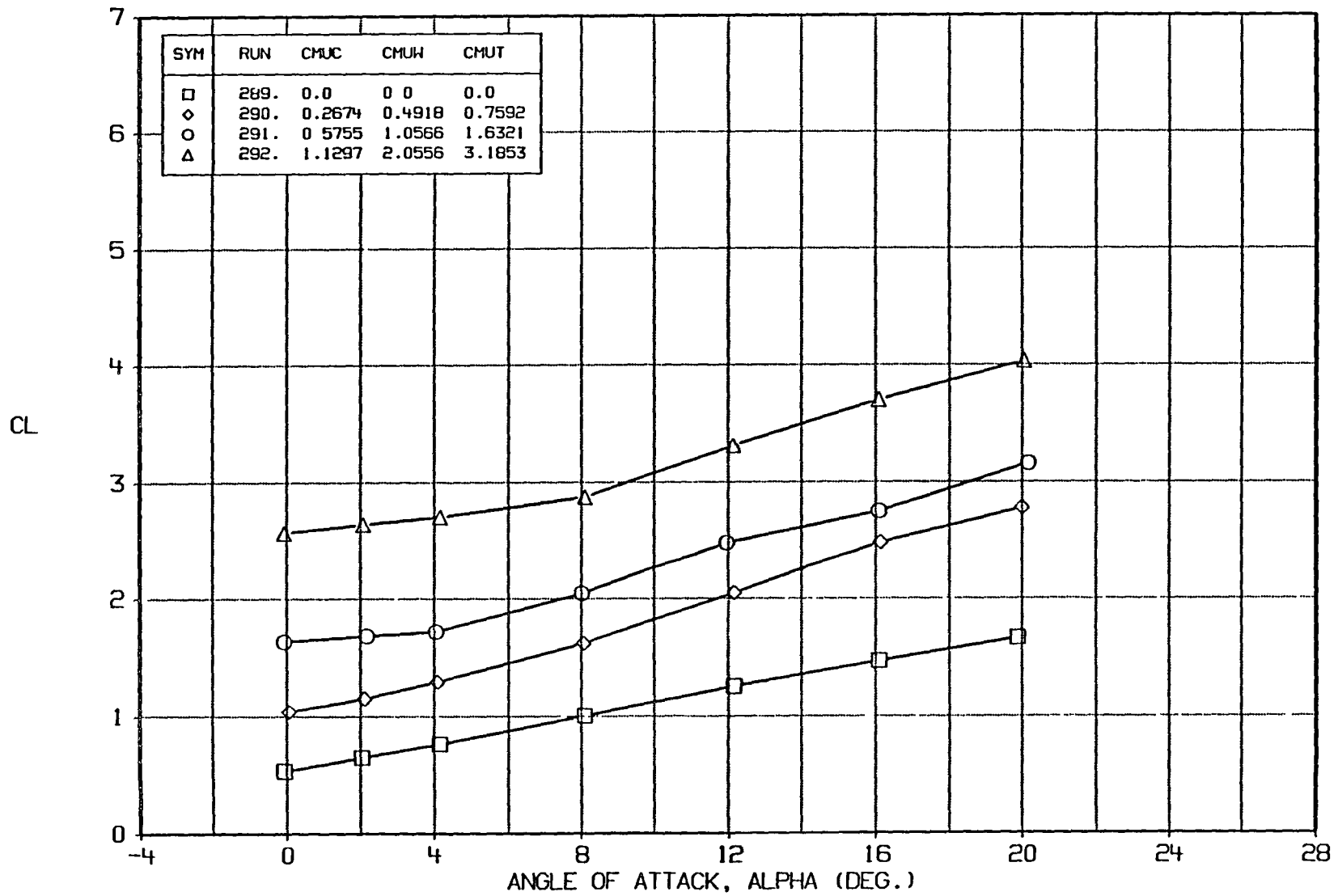


FIGURE A21a BASIC DATA EFFECT OF CMU  
CONFIG. BC1W6V, DELF=45, (BN/B)C=1, (BN/B)W=0.25

A-131

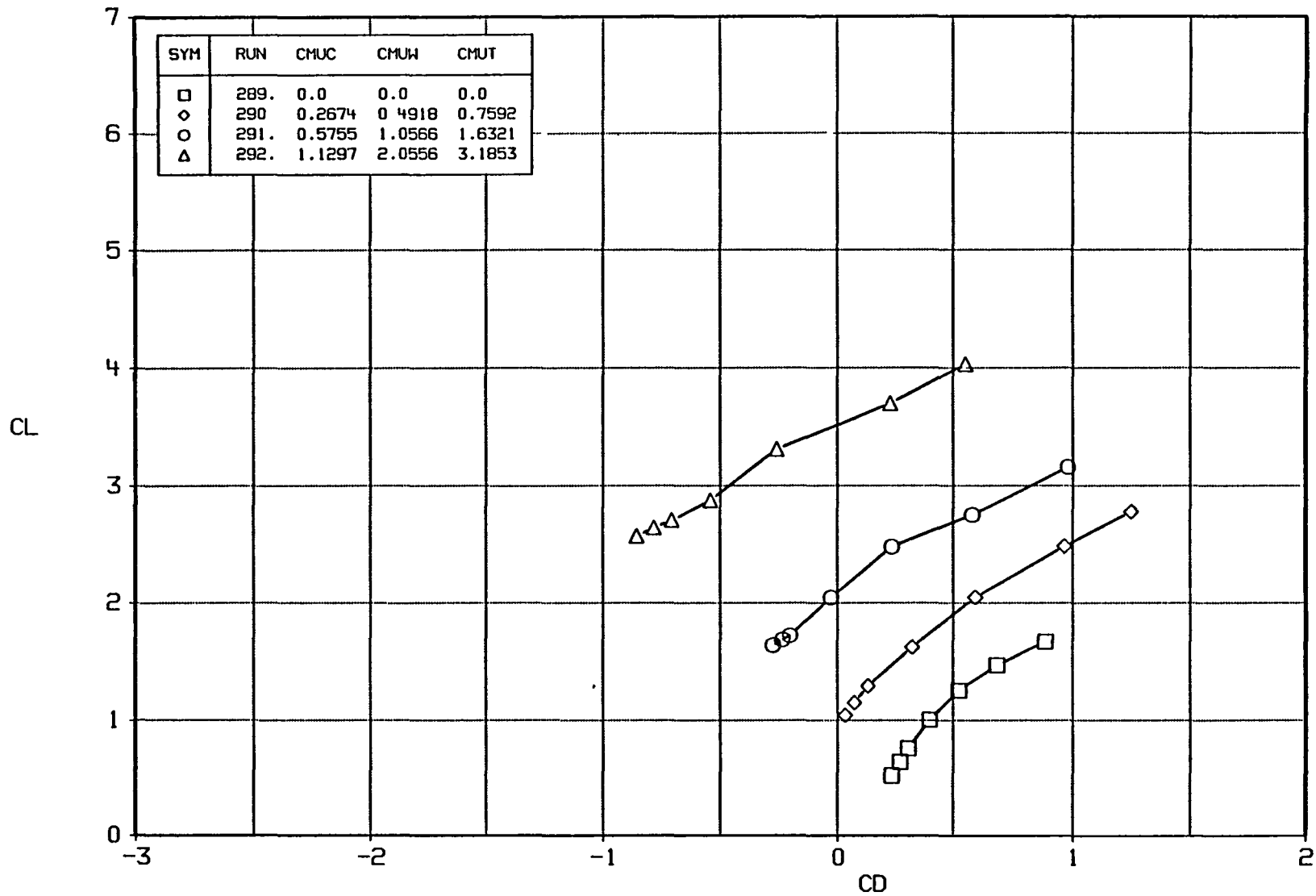


FIGURE A21b BASIC DATA EFFECT OF CMU  
CONFIG. BC1W6V, DELF=45, (BN/B)C=1, (BN/B)W=0.25

A-132

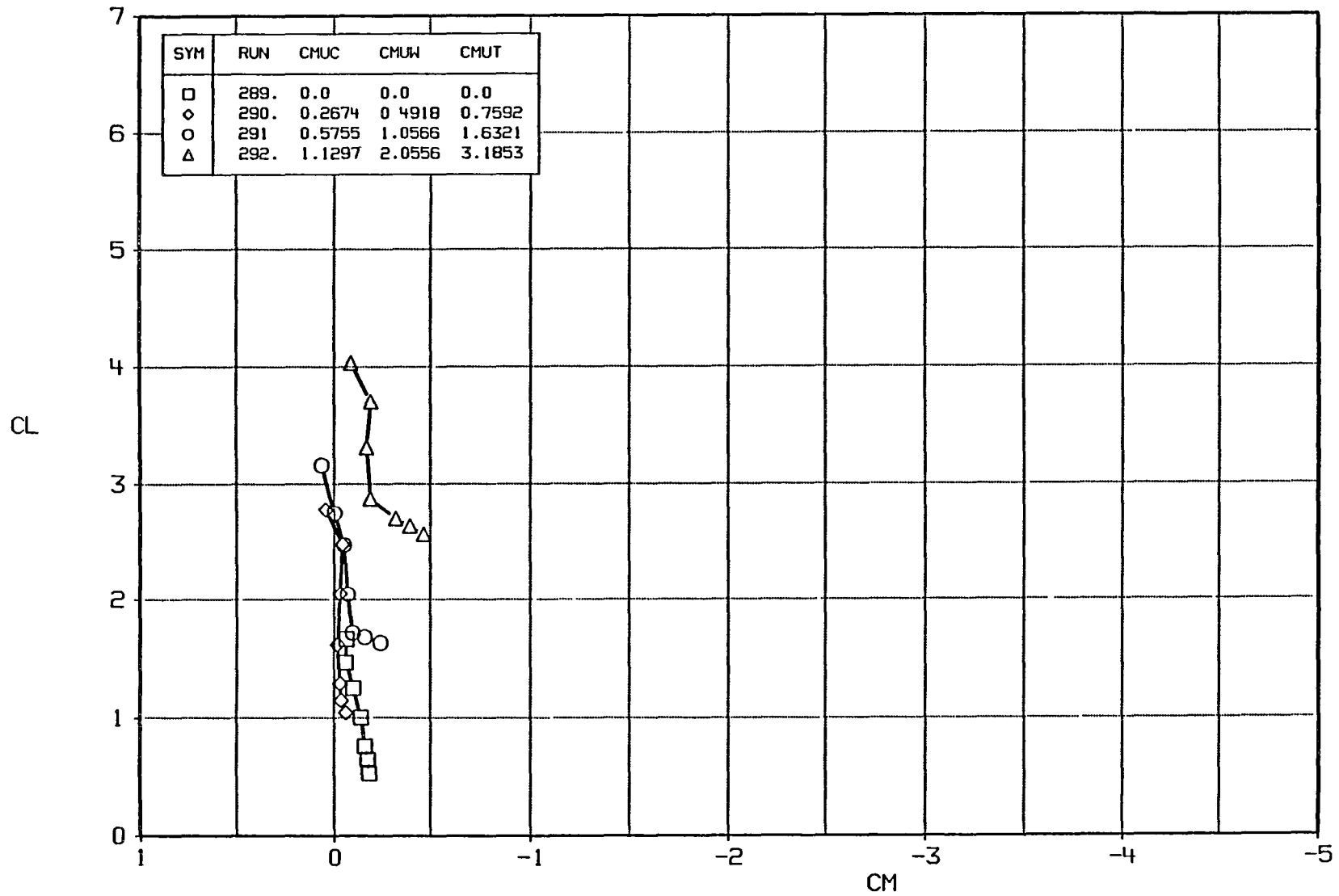


FIGURE A21c BASIC DATA EFFECT OF CMU  
CONFIG. BC1W6V, DELF=45, (BN/B)C=1, (BN/B)W=0.25

A-133

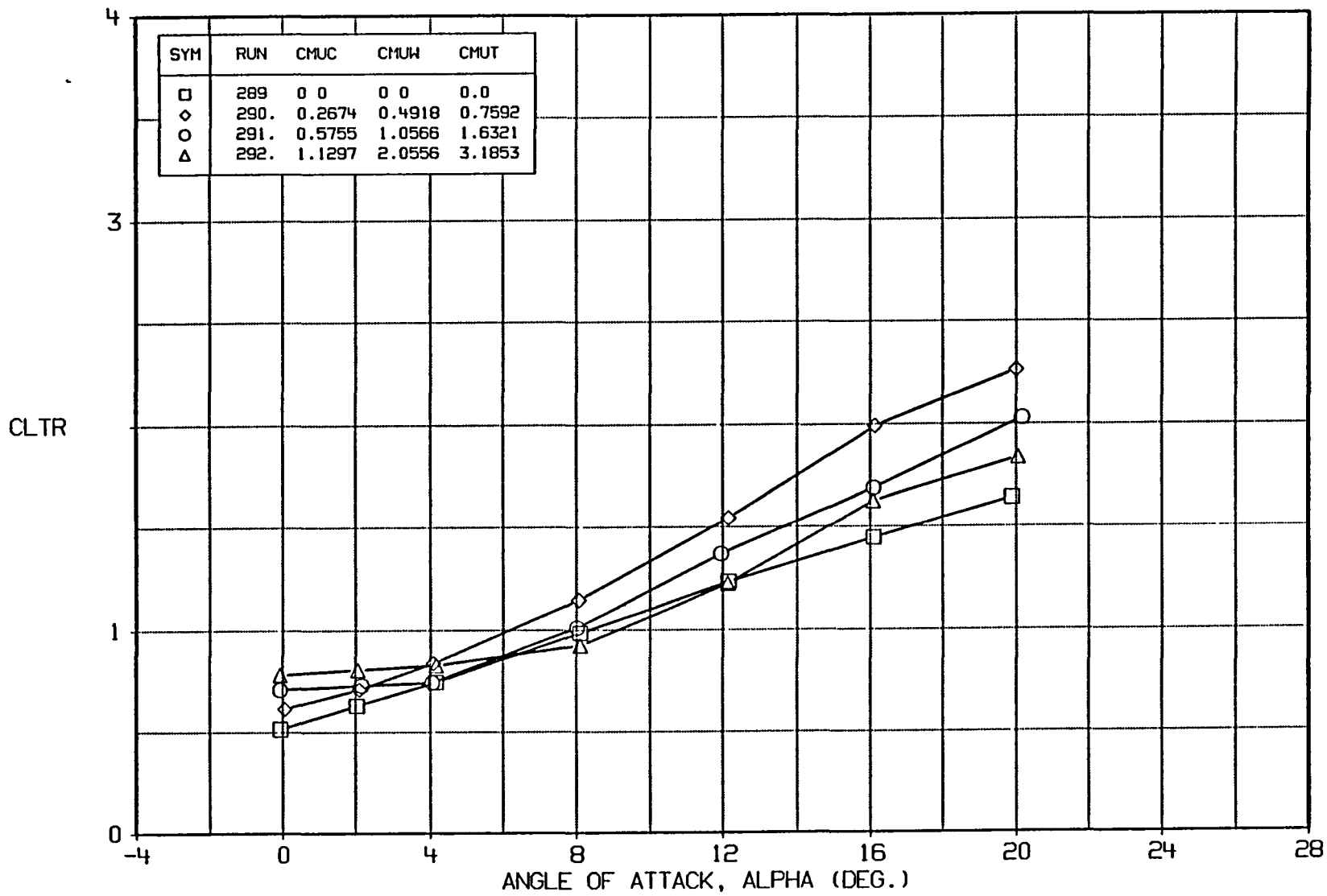


FIGURE A21d BASIC DATA EFFECT OF CMU  
CONFIG. BC1W6V, DELF=45, (BN/B)C=1, (BN/B)W=0.25

A-134

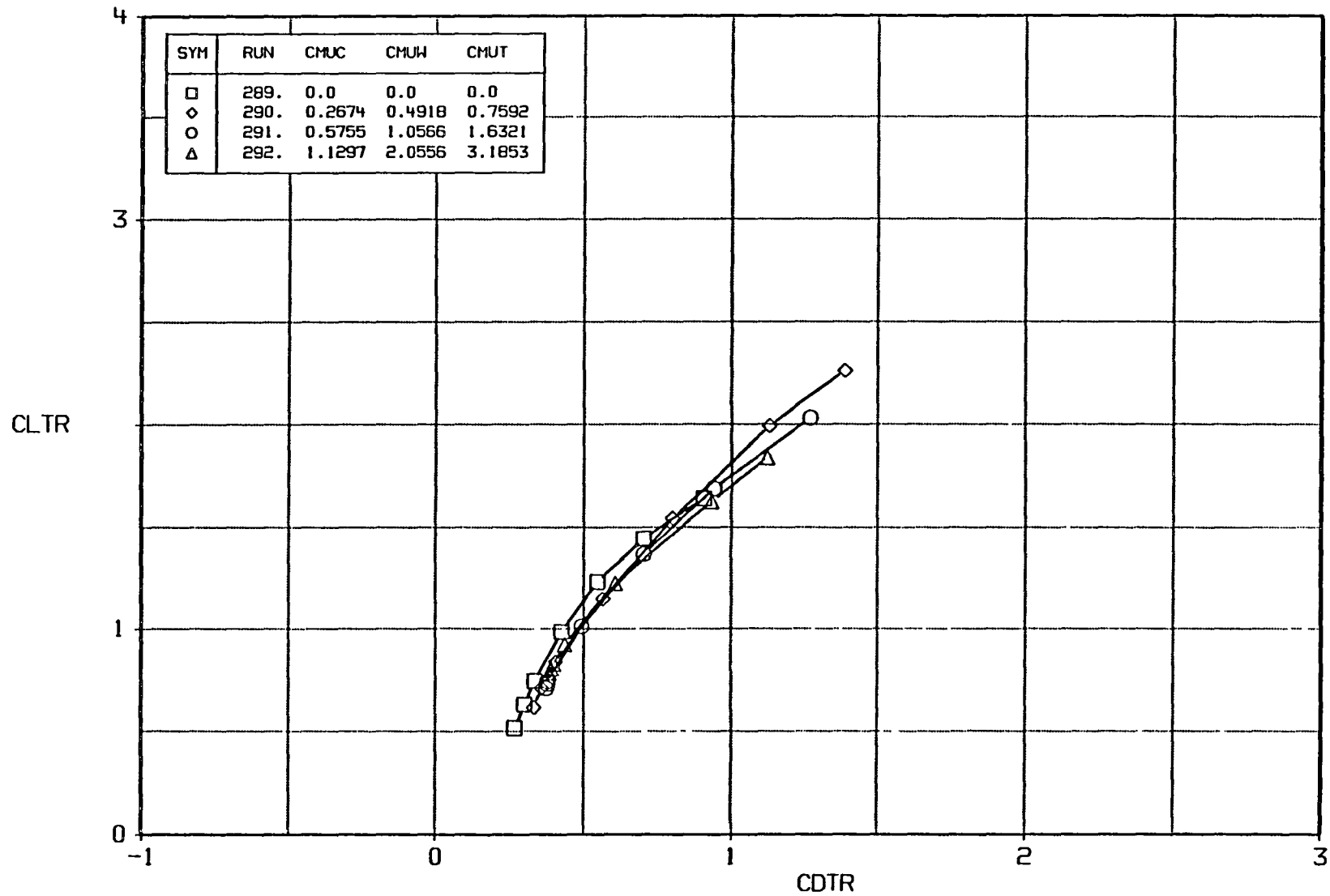


FIGURE A21e BASIC DATA EFFECT OF CMU  
CONFIG. BC1W6V, DELF=45, (BN/B)C=1, (BN/B)W=0.25



A-135

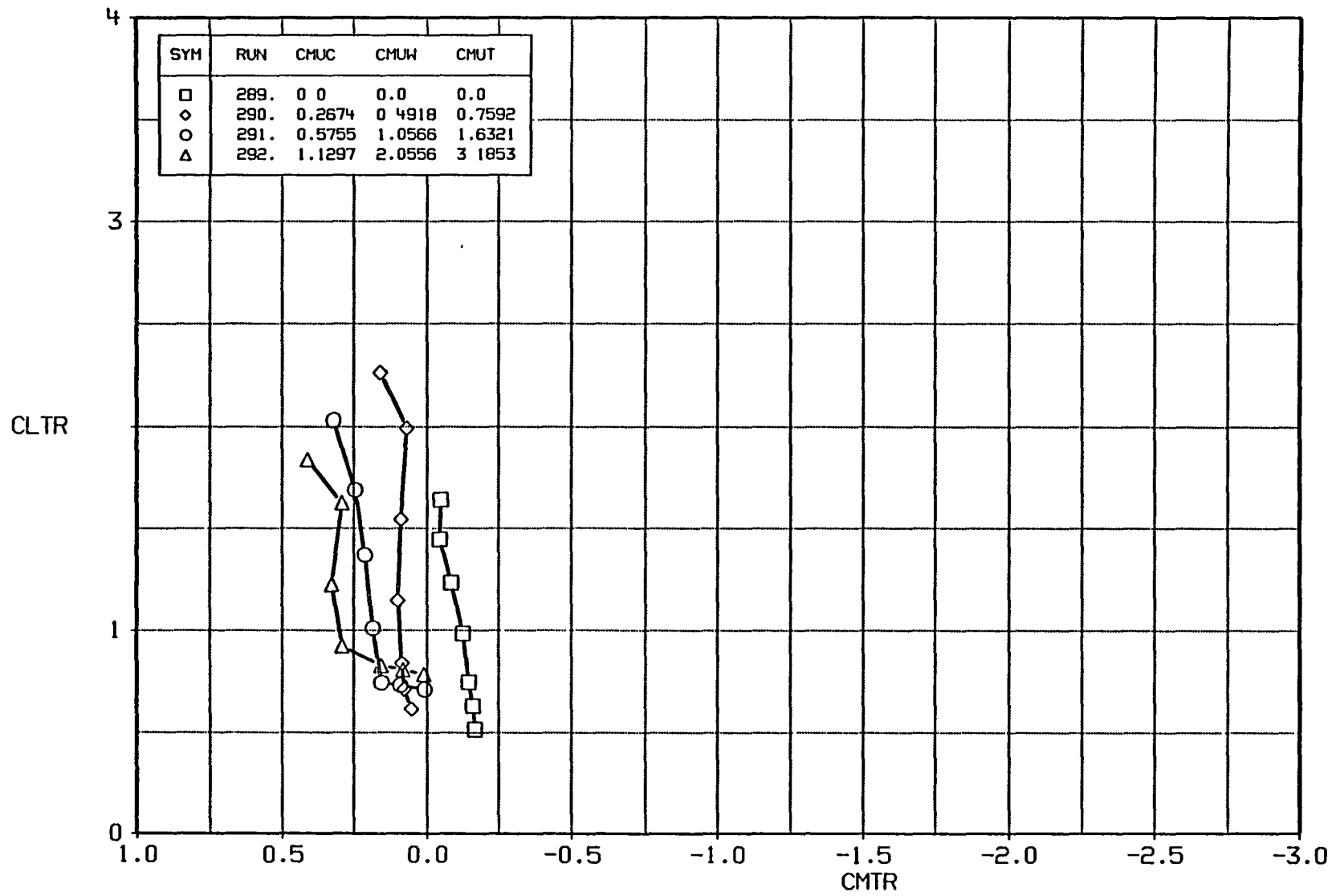


FIGURE A21f BASIC DATA EFFECT OF CMU  
CONFIG. BC1W6V, DELF=45, (BN/B)C=1, (BN/B)W=0.25

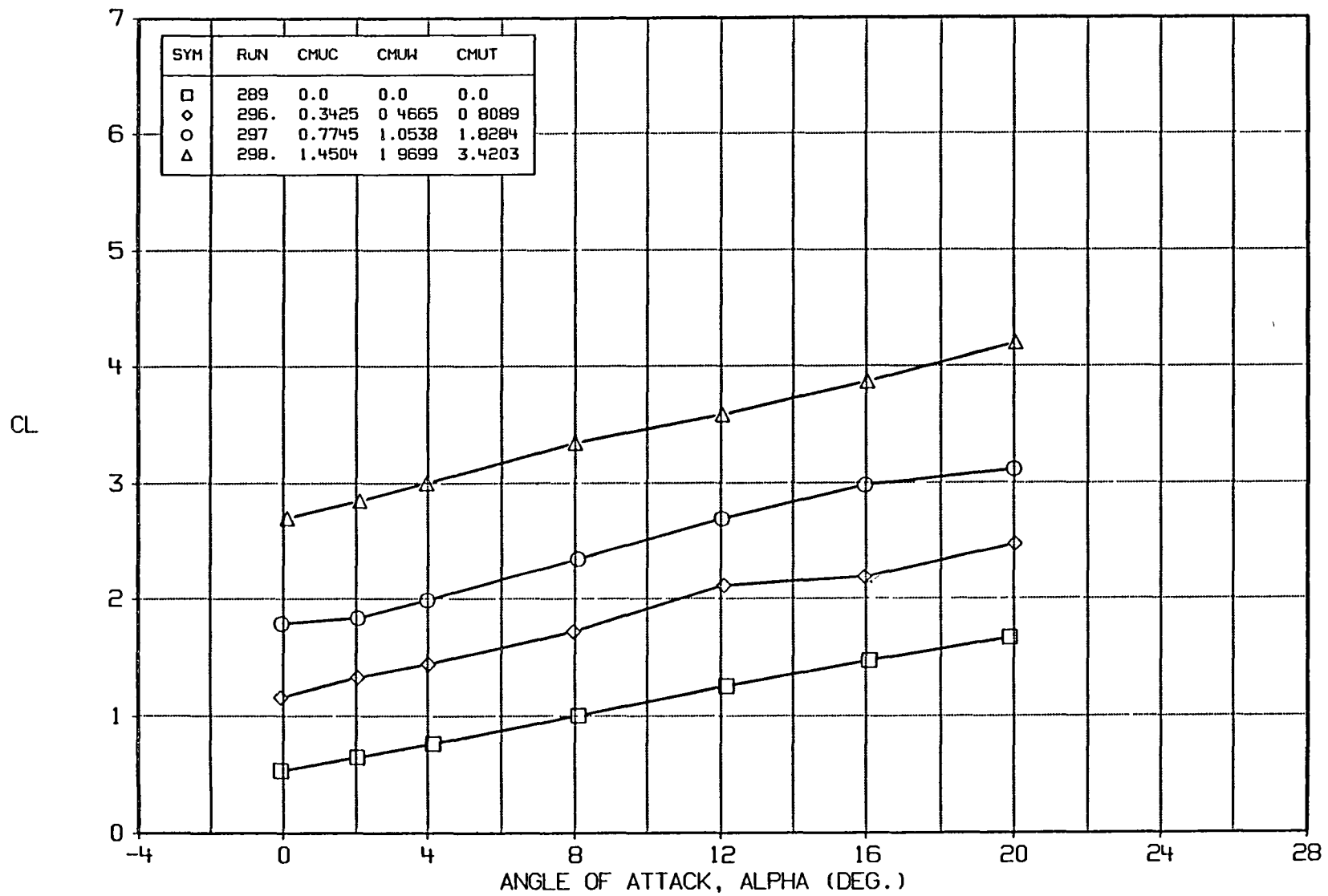


FIGURE A22a BASIC DATA EFFECT OF CMU  
 CONFIG. BC1W6V DELF=45  $(BN/B)_C=0.5$   $(BN/B)_W=0.25$

A-137

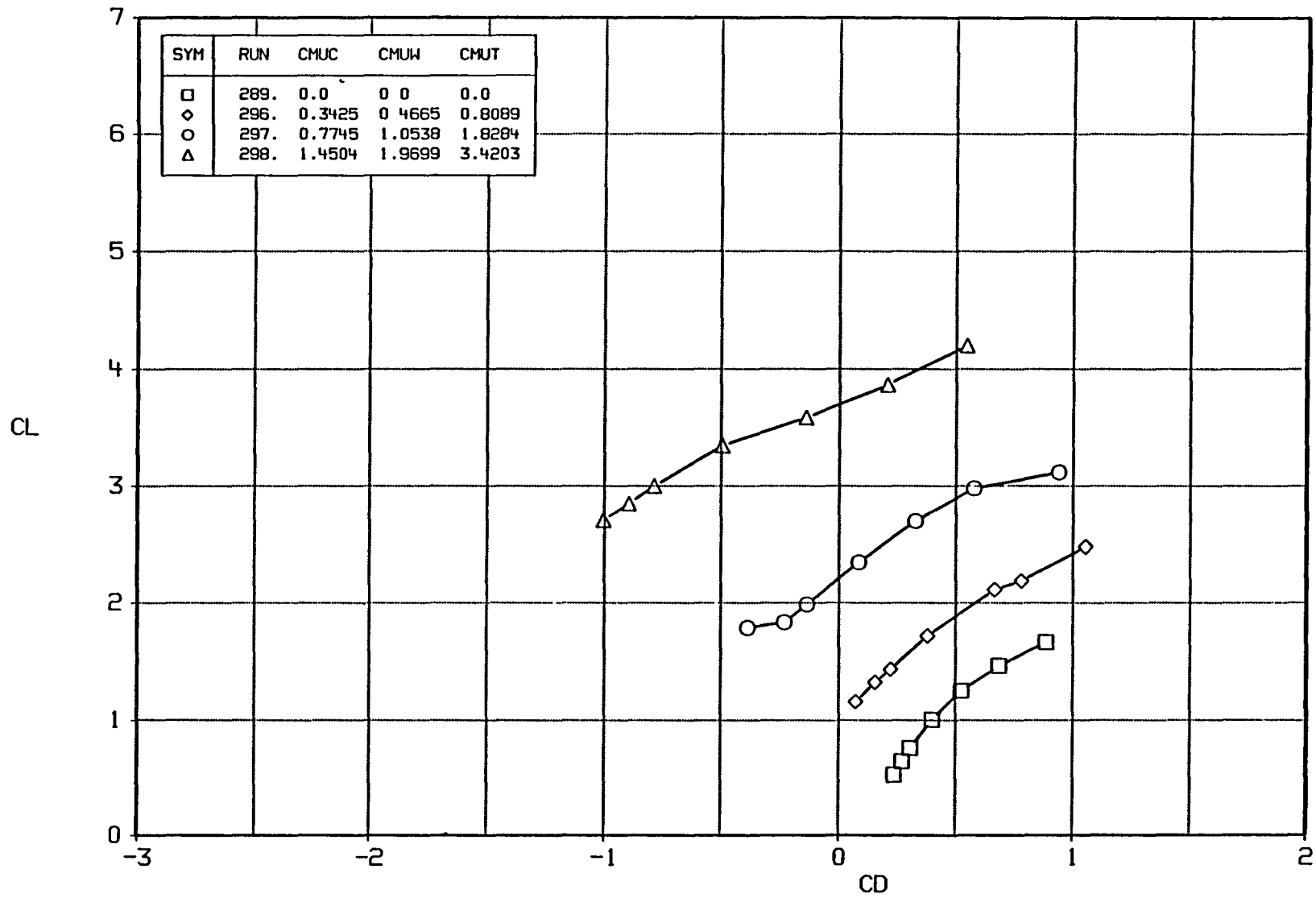


FIGURE A22b BASIC DATA EFFECT OF CMU  
 CONFIG. BCW6V DELF=45  $(BN/B)_C=0.5$   $(BN/B)_W=0.25$

A-138

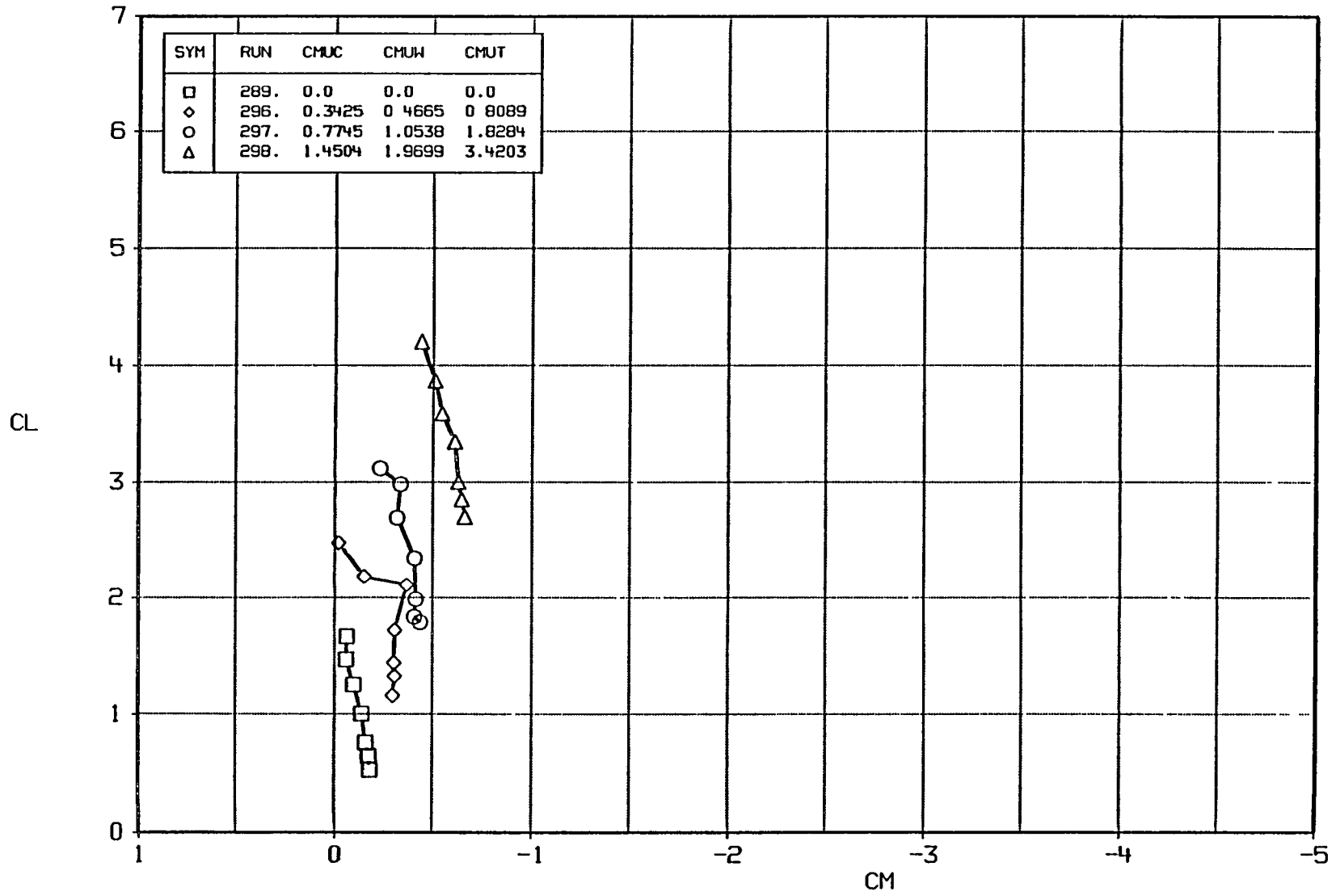


FIGURE A22c BASIC DATA EFFECT OF CMU  
CONFIG. BC1W6V DELF=45  $(BN/B)_C=0.5$   $(BN/B)_W=0.25$

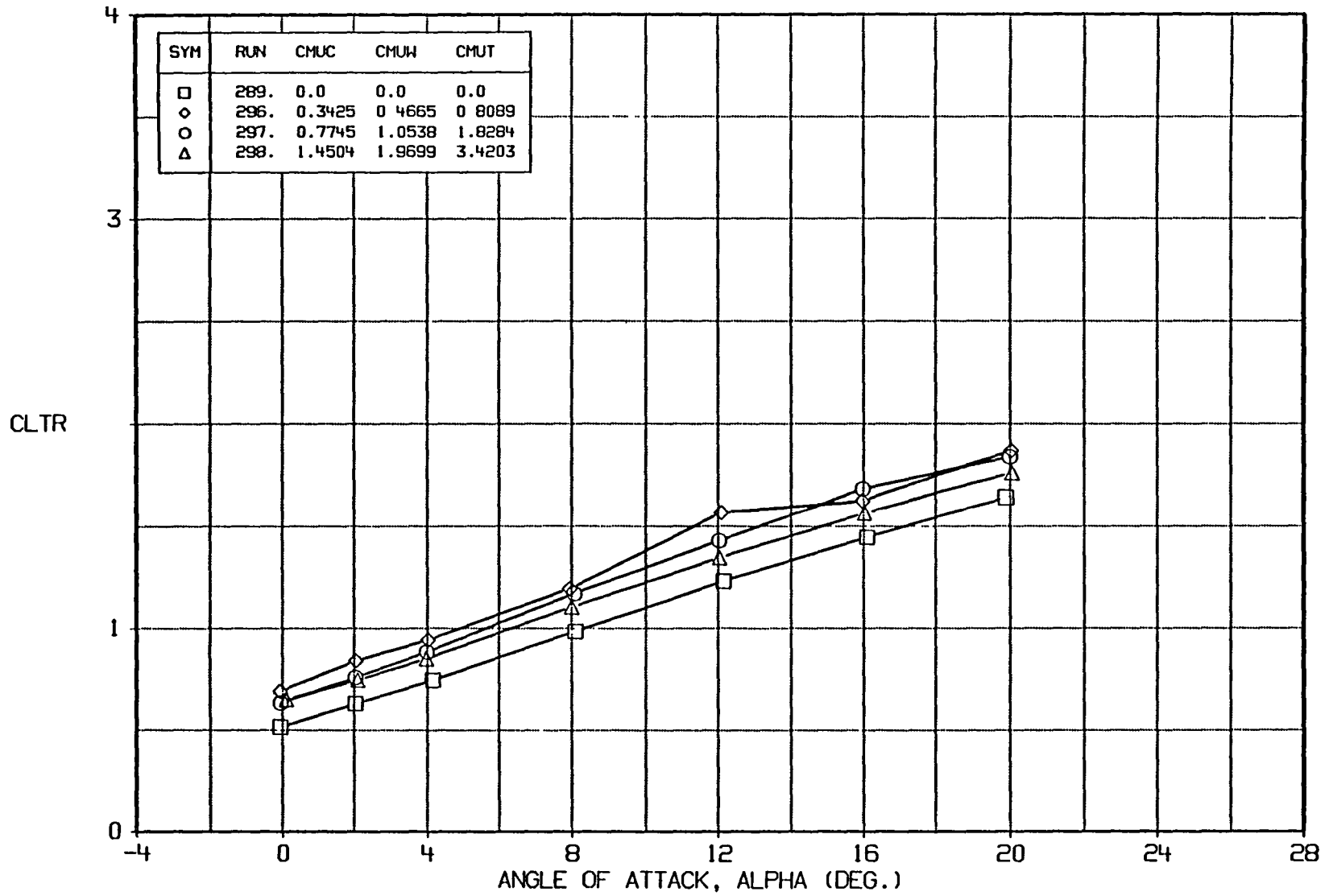


FIGURE A22d BASIC DATA EFFECT OF CMU  
 CONFIG. BC1W6V DELF=45  $(BN/B)_C=0.5$   $(BN/B)_W=0.25$

A-140

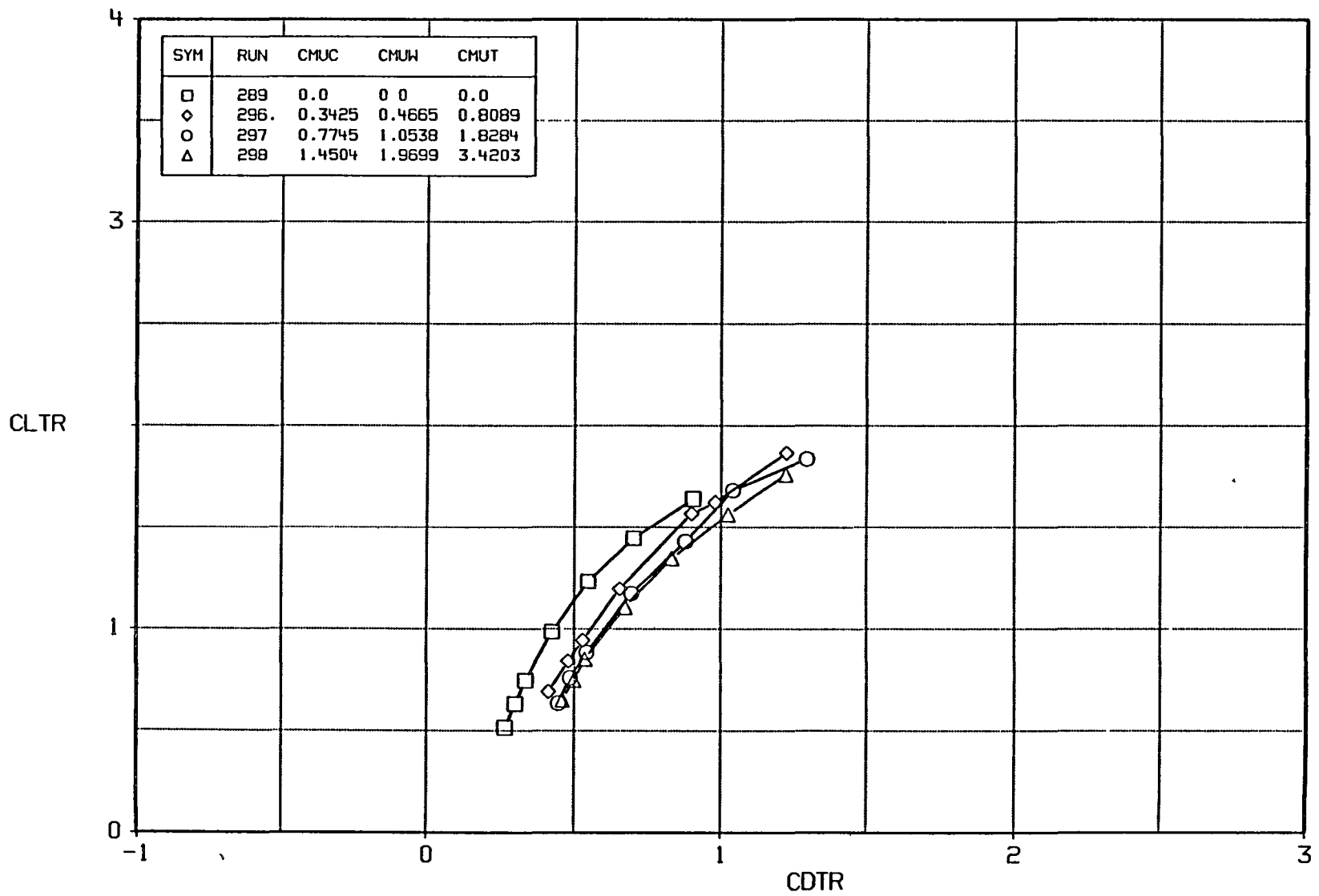


FIGURE A22e BASIC DATA EFFECT OF CMU  
CONFIG. BC1W6V DELF=45  $(BN/B)_C=0.5$   $(BN/B)_W=0.25$

A-141

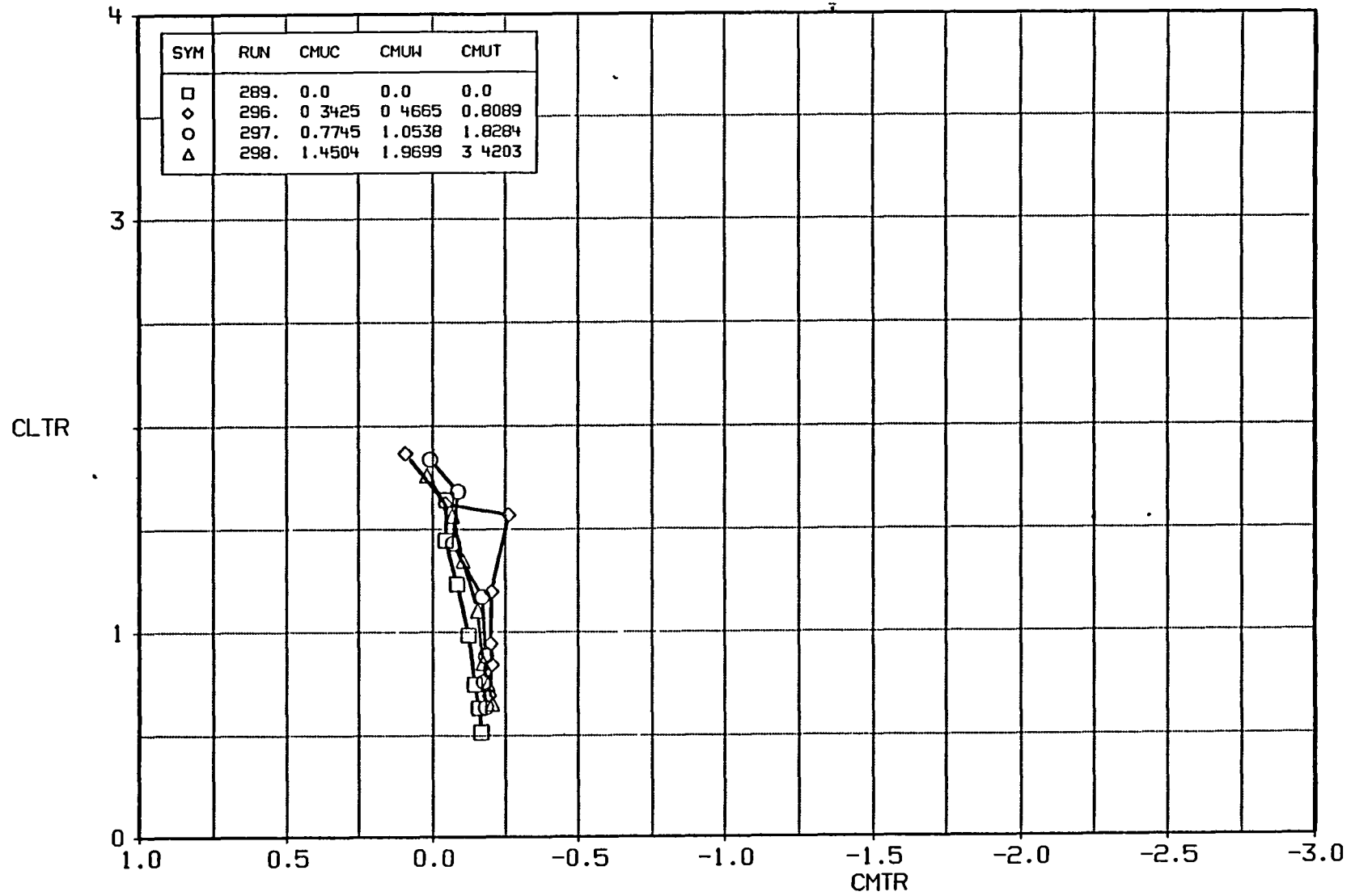


FIGURE A22f BASIC DATA EFFECT OF CMU  
CONFIG. BC1W6V DELF=45  $(BN/B)_C=0.5$   $(BN/B)_W=0.25$

A-142

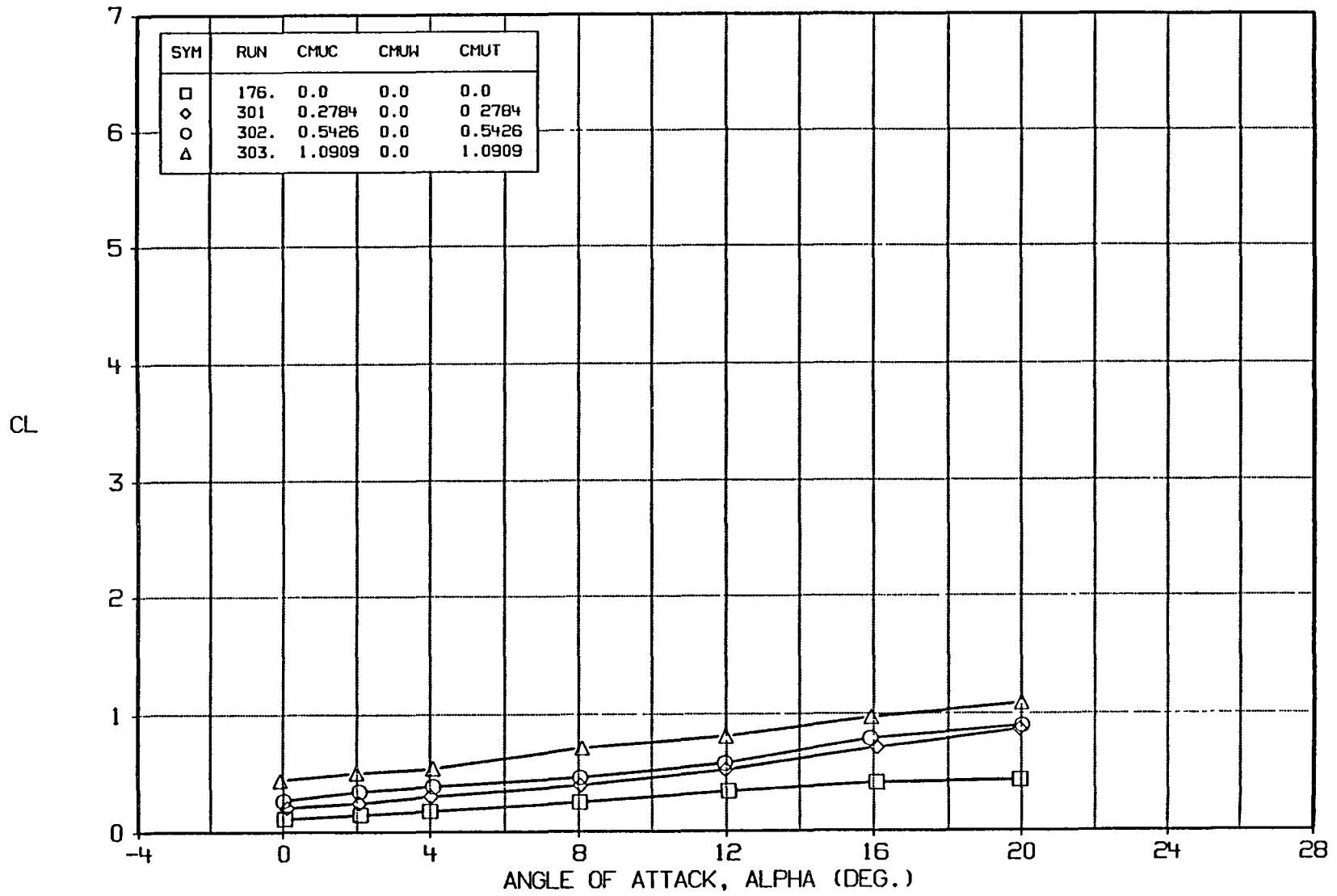


FIGURE A23a BASIC DATA EFFECT OF CMU  
CONFIG. BC1V, DELF=45, (BN/B)C=0.5



A-143

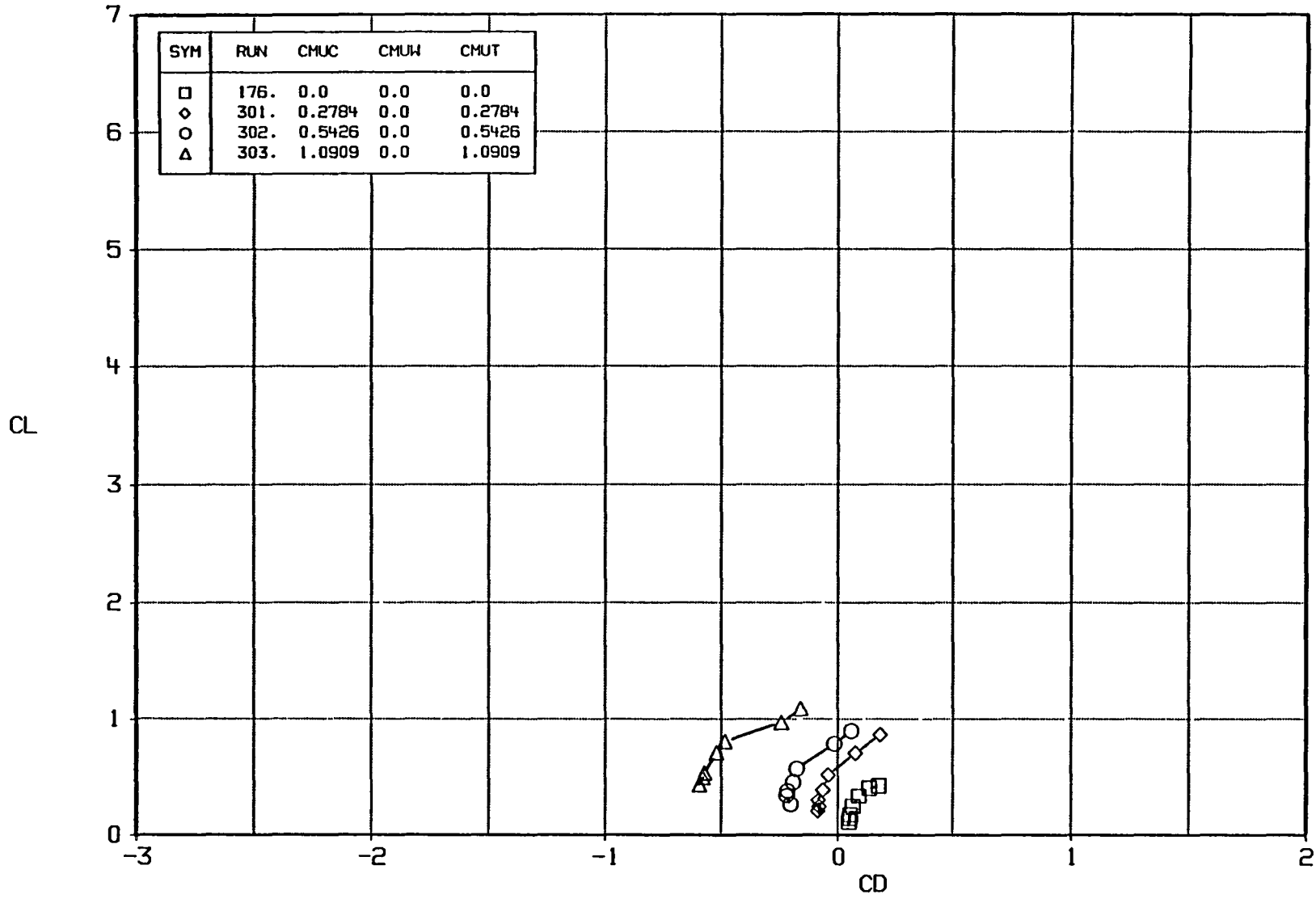


FIGURE A23b BASIC DATA EFFECT OF CMU  
CONFIG. BC1V, DELF=45, (BN/B)C=0.5,

A-144

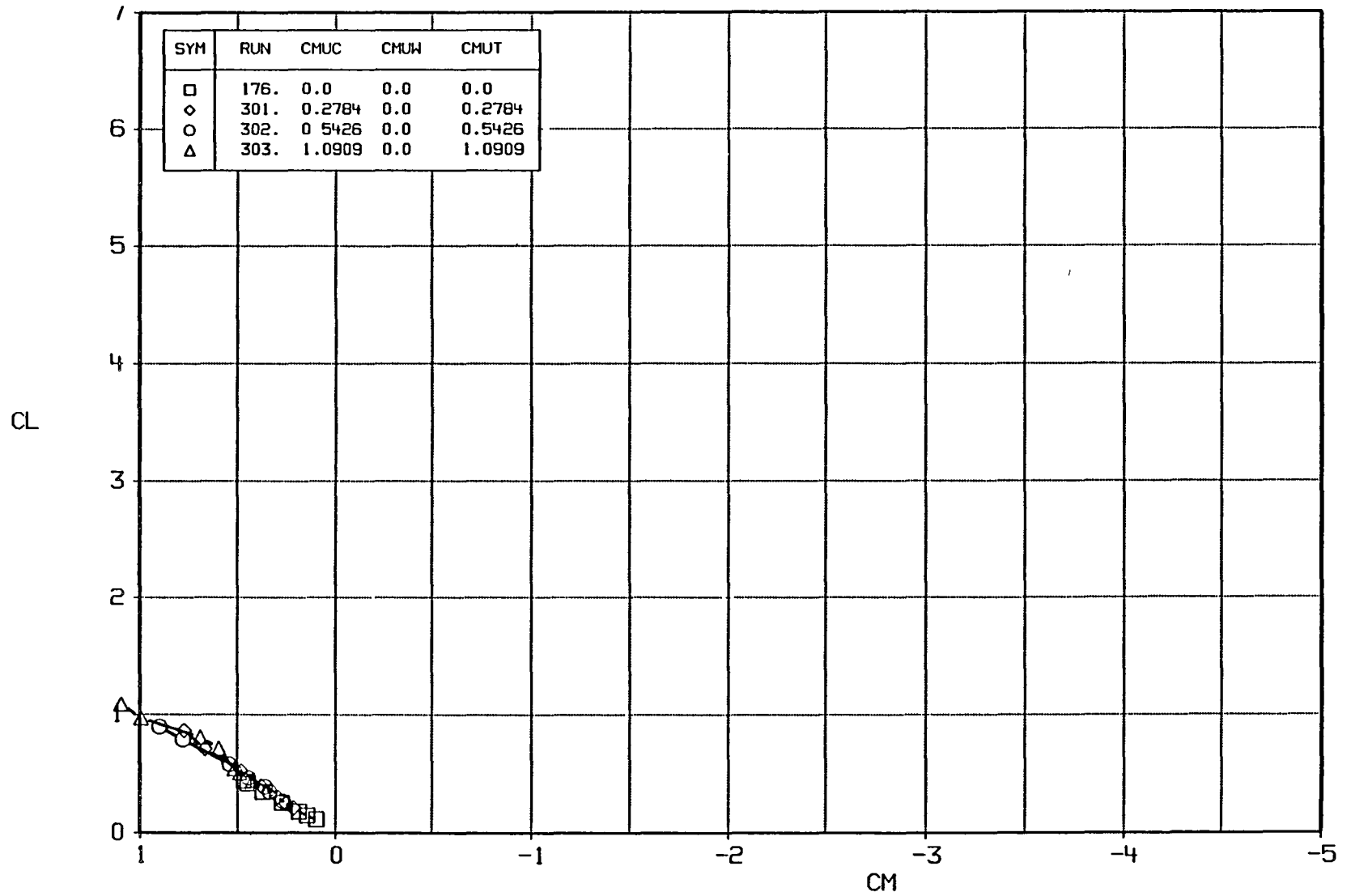


FIGURE A23c BASIC DATA EFFECT OF CMU  
CONFIG. BC1V, DELF=45, (BN/B)C=0.5

A-145

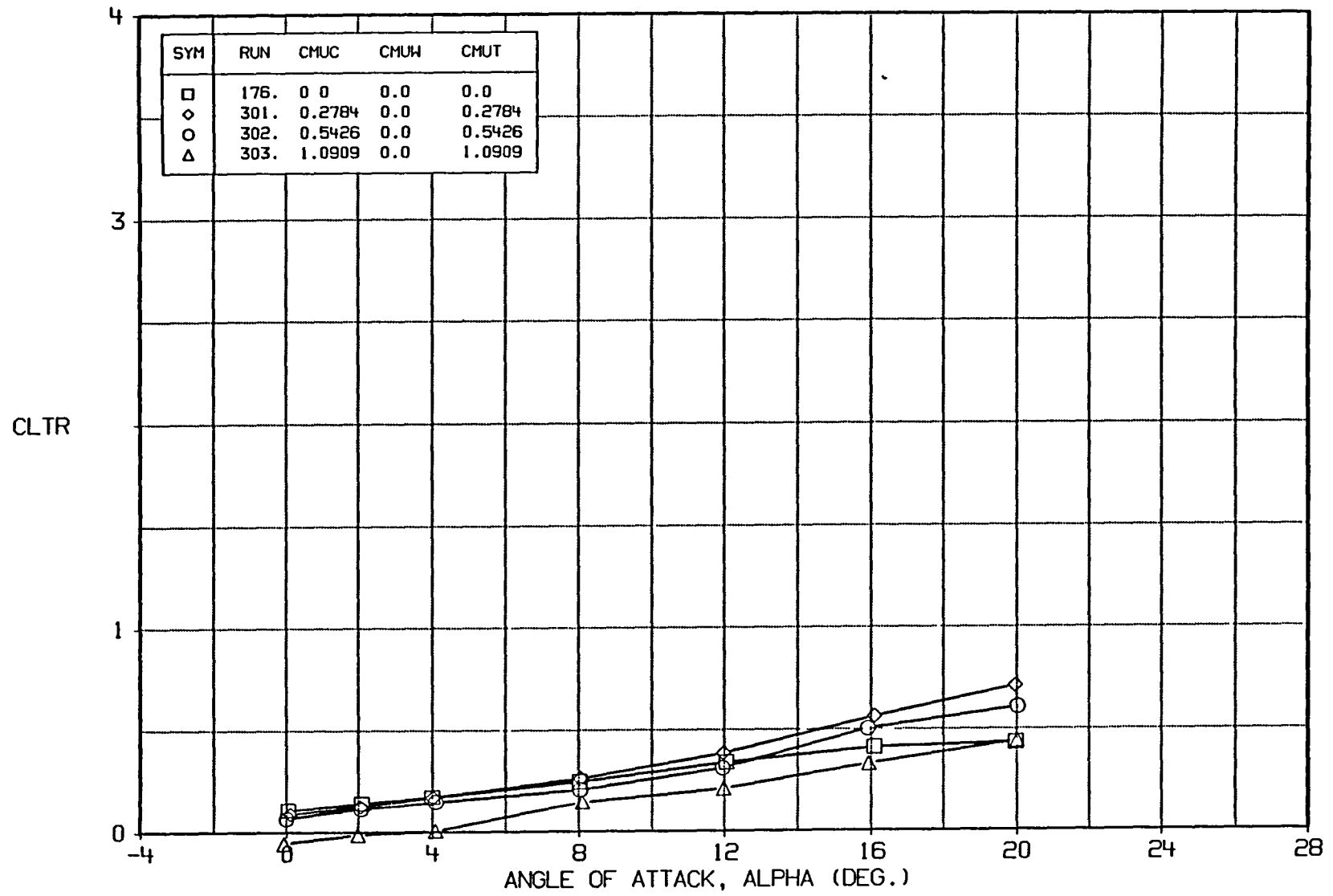


FIGURE A23d BASIC DATA EFFECT OF CMU  
CONFIG. BC1V, DELF=45, (BN/B)C=0.5, (BN/B)W=0

A-146

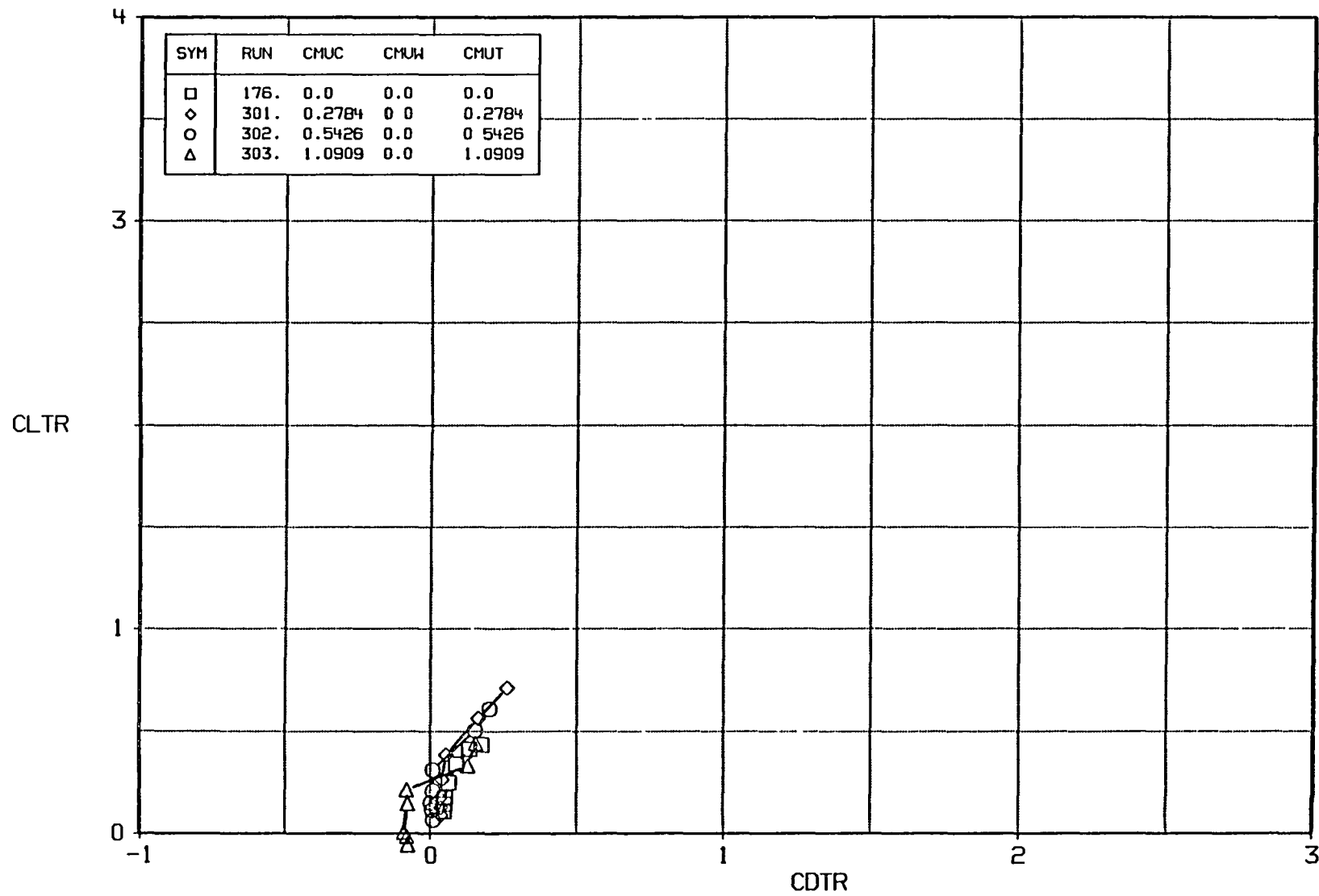


FIGURE A23e BASIC DATA EFFECT OF CMU  
CONFIG. BC1V, DELF=45, (BN/B)C=0.5

A-147

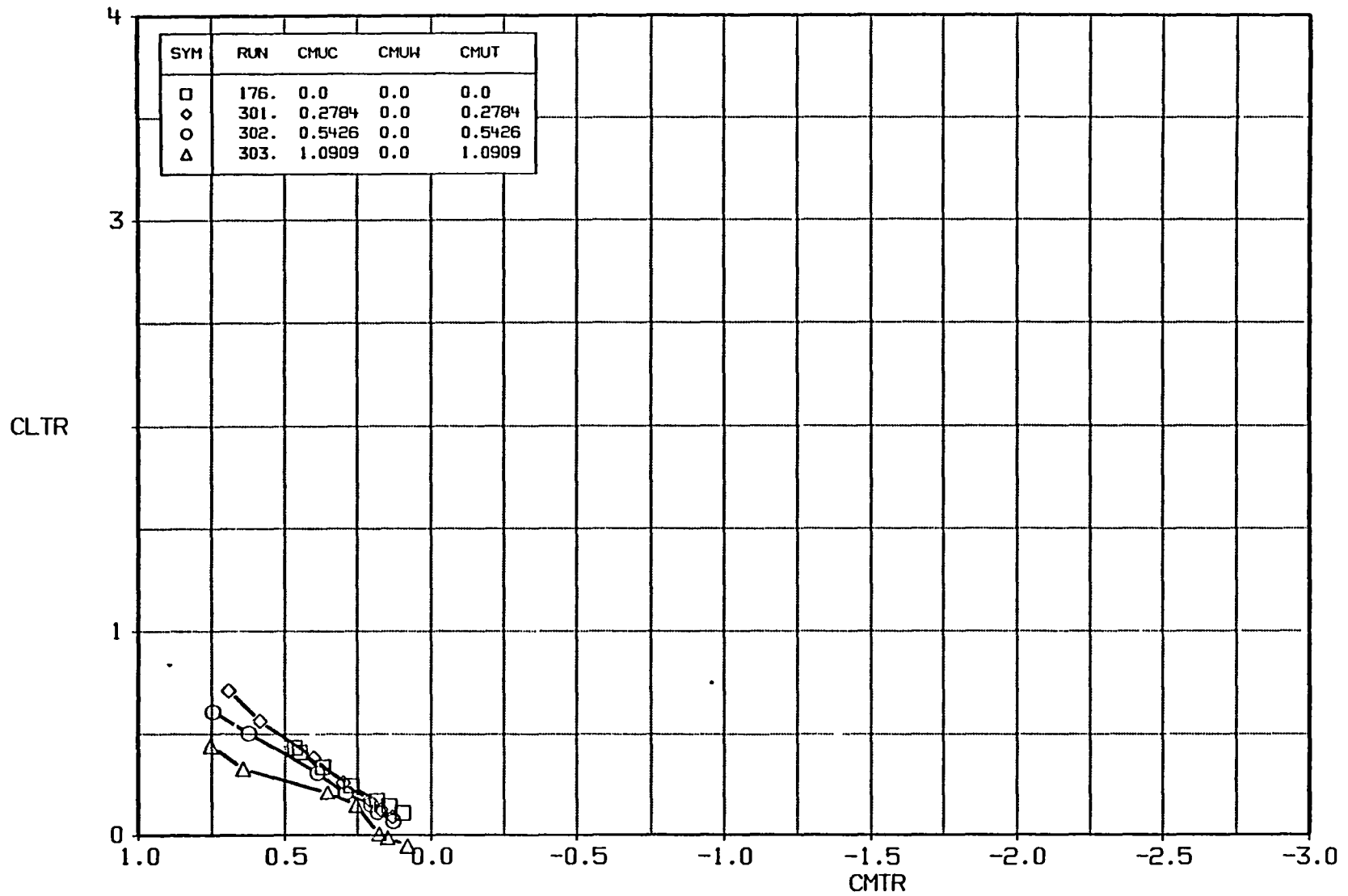


FIGURE A23f BASIC DATA EFFECT OF CMU  
CONFIG. BC1V, DELF=45, (BN/B)C=0.5,

A-148

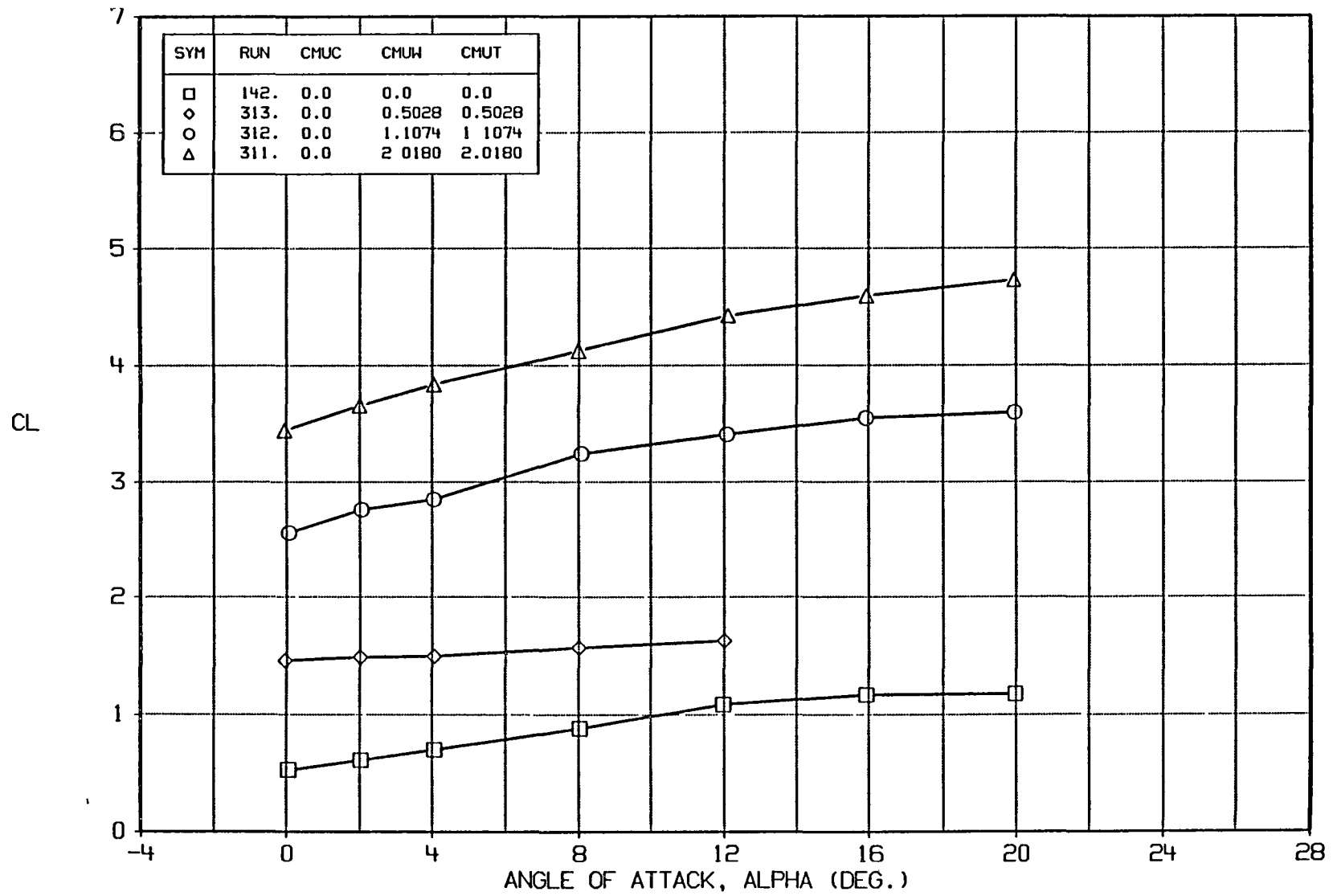


FIGURE A24a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

A-149

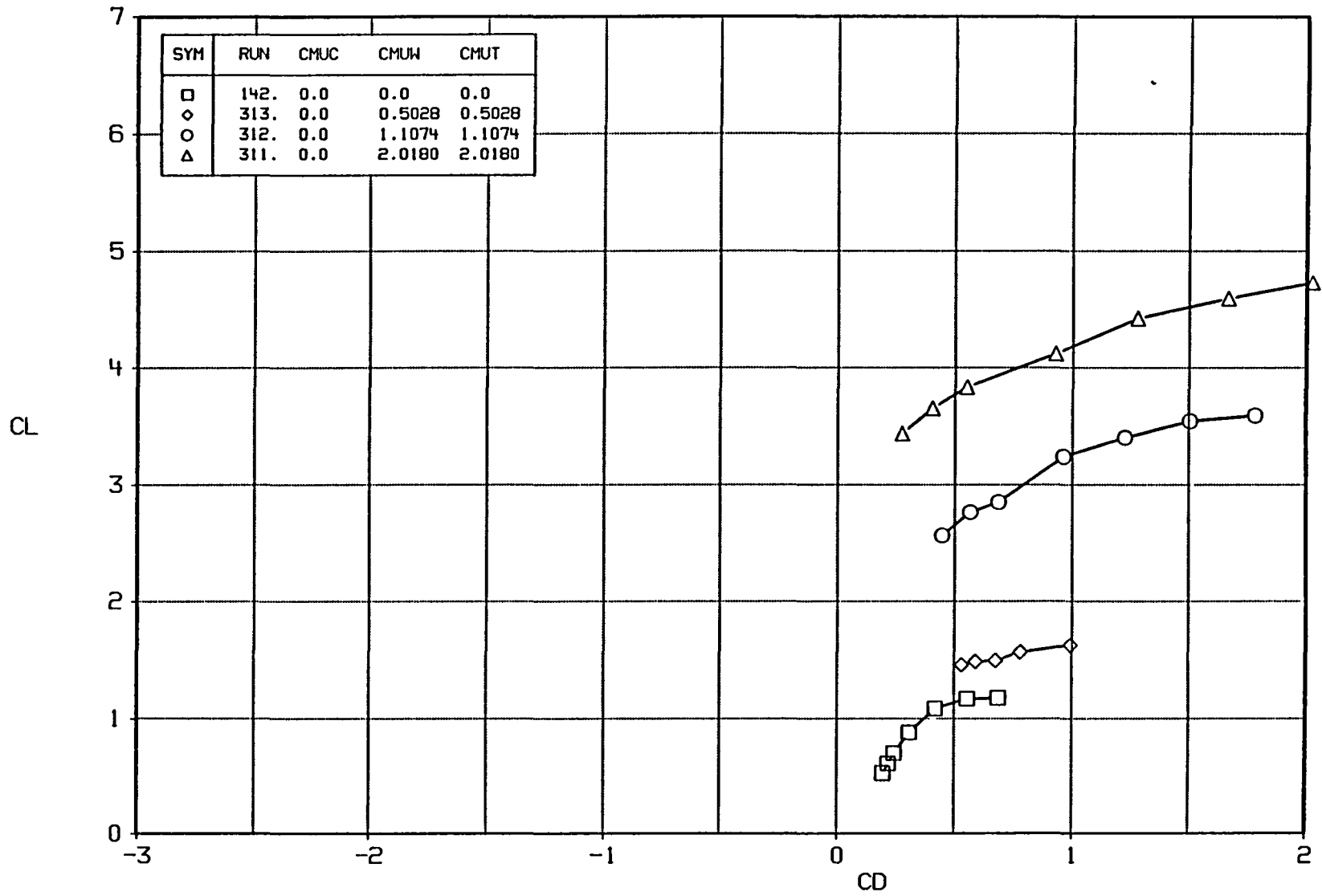


FIGURE A24b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

A-150

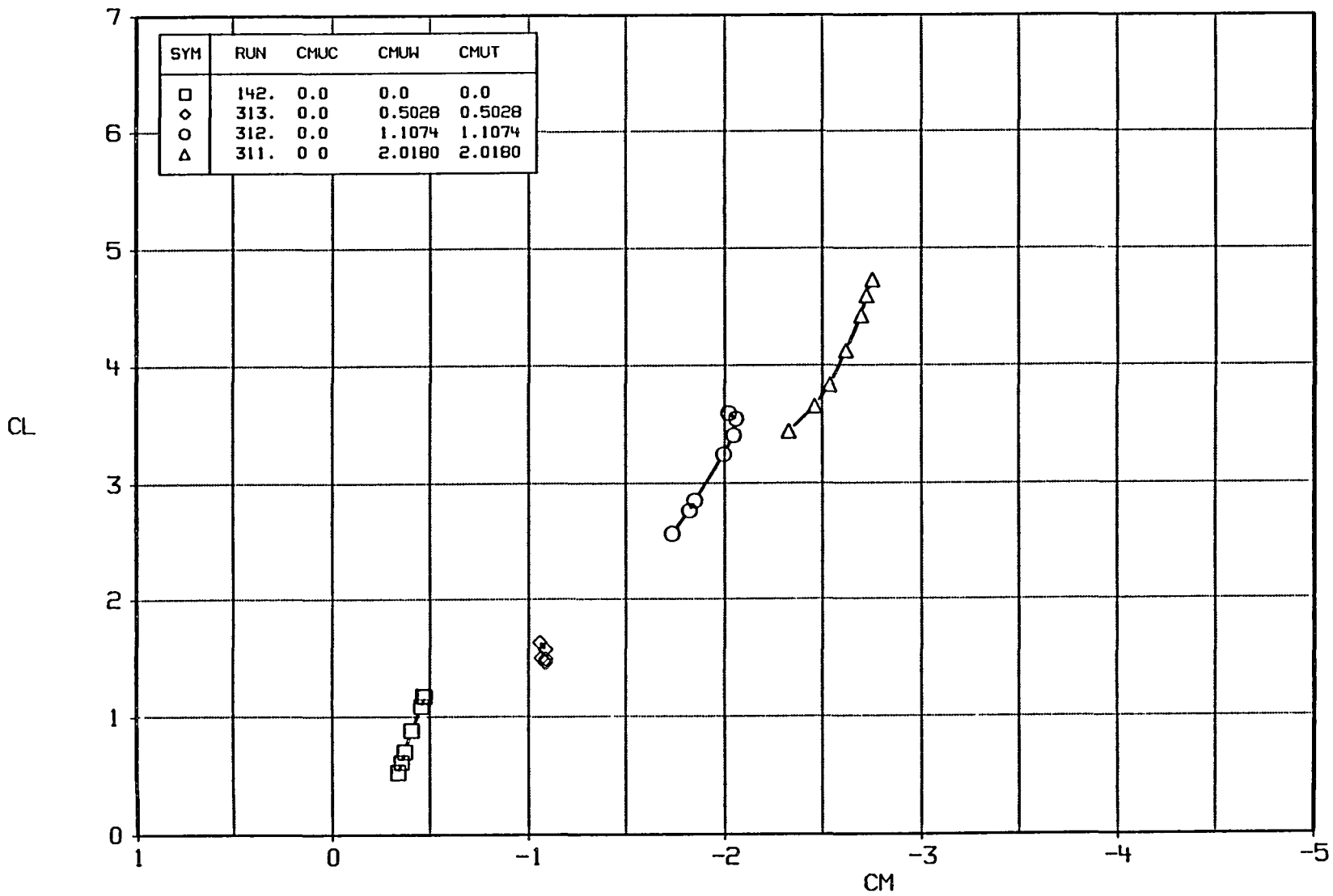


FIGURE A24c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1



A-151

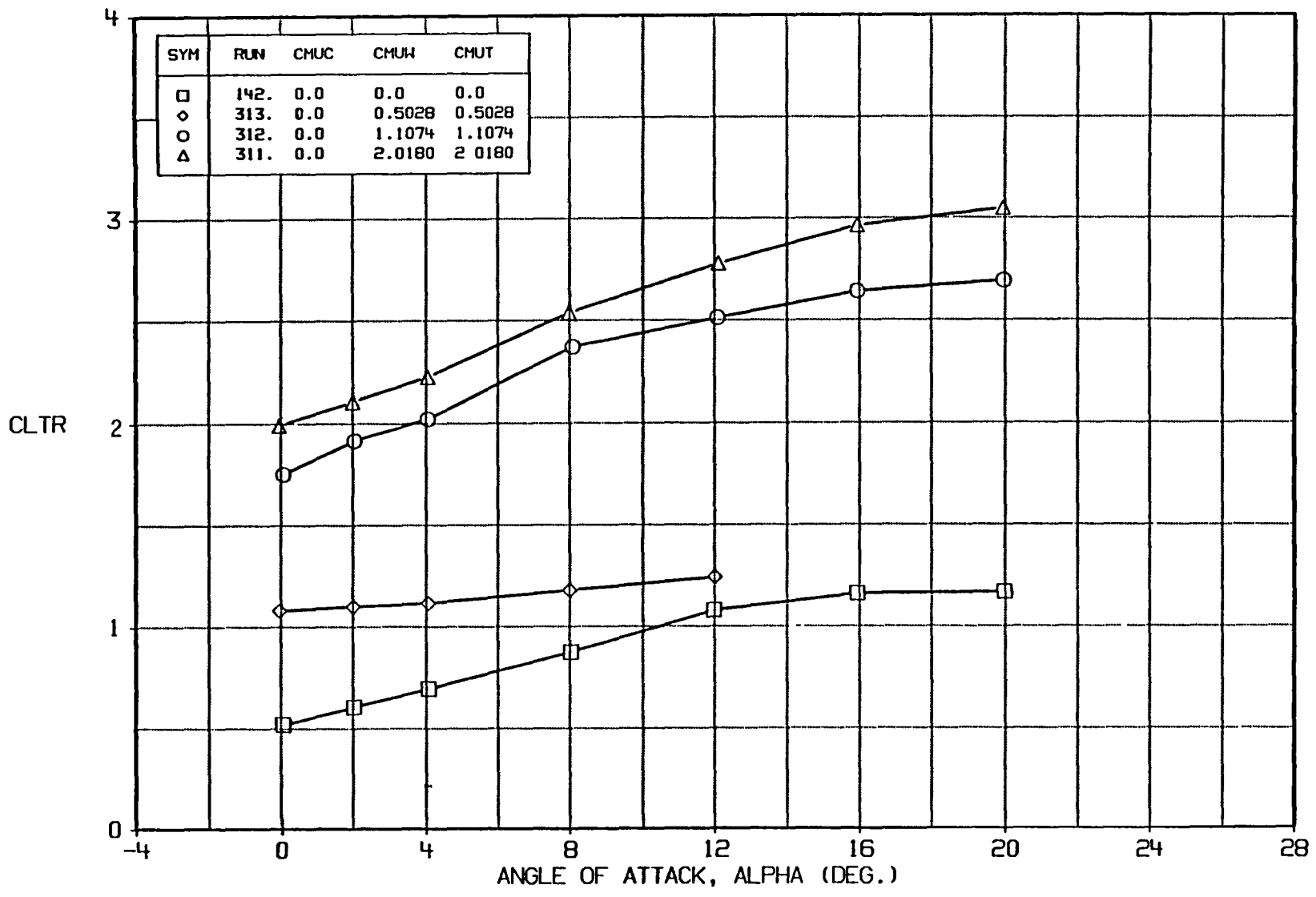


FIGURE A24d BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

A-152

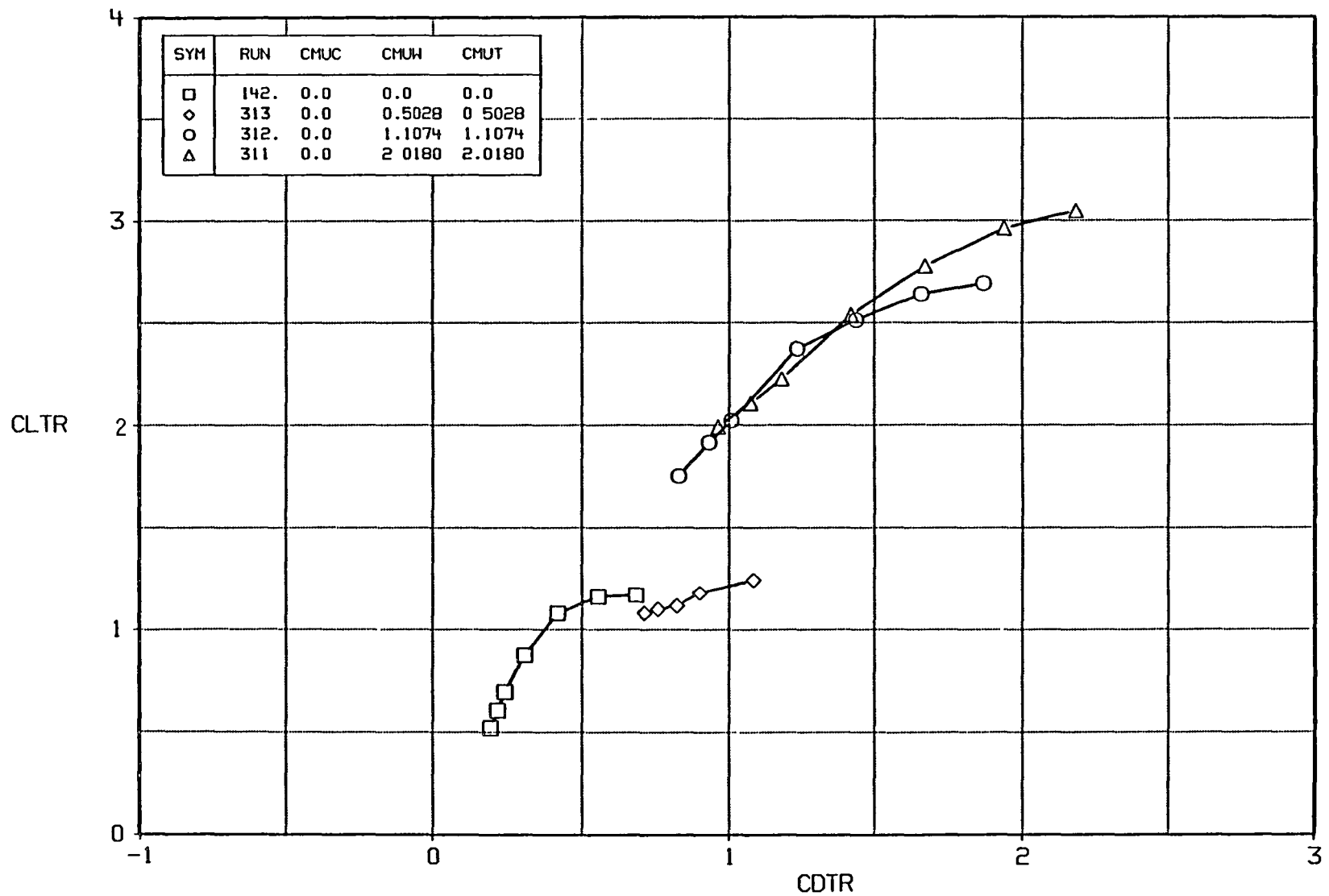


FIGURE A24e BASIC DATA EFFECT OF CMU CONFIGURATION BW6V, DELF=45, BN/B=1

A-153

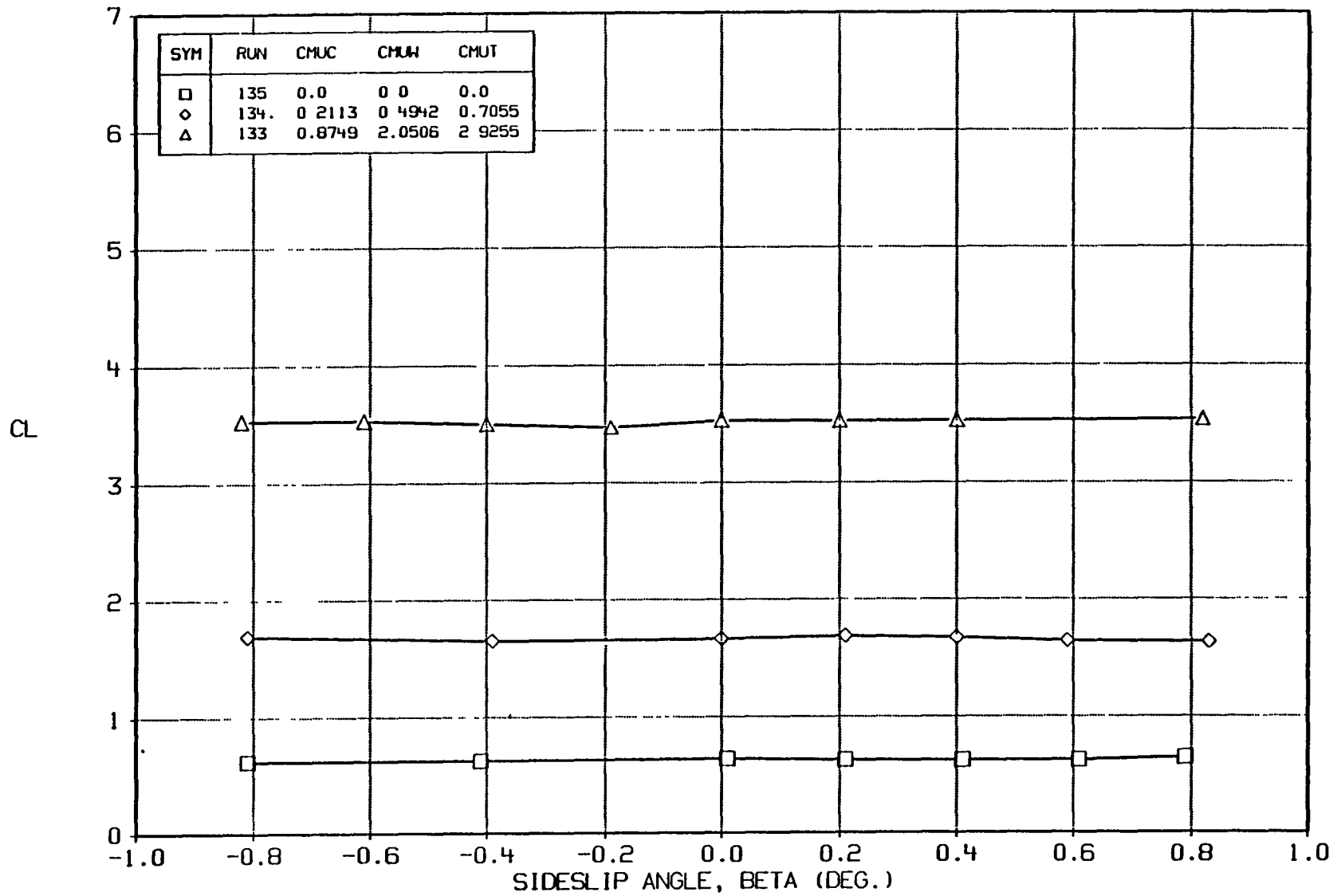


FIGURE A25a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=1.0, DELF=45,  $\alpha = 2$ .

A-154

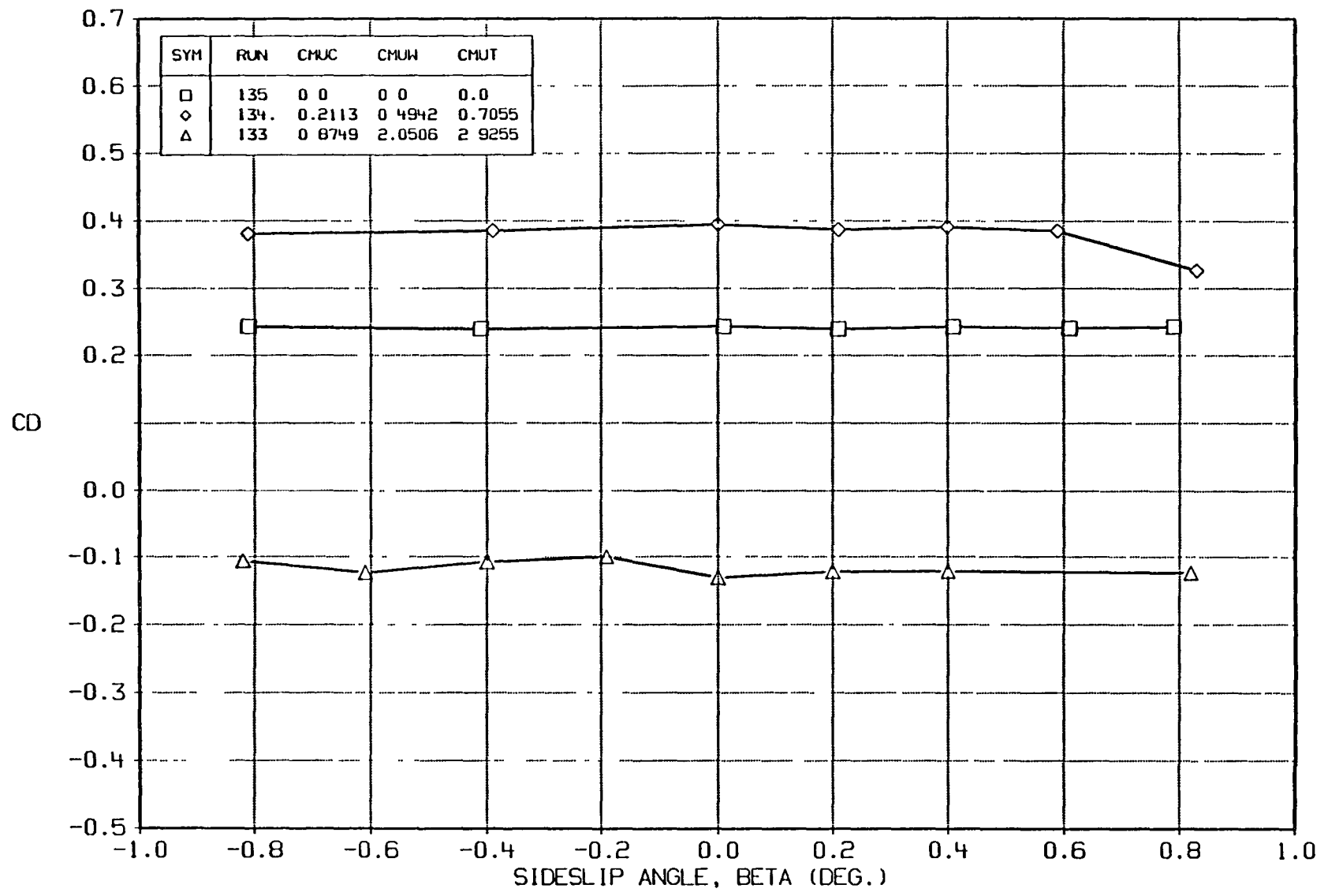


FIGURE A25b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=1.0, DELF=45,  $\alpha = 2$  .

A-155

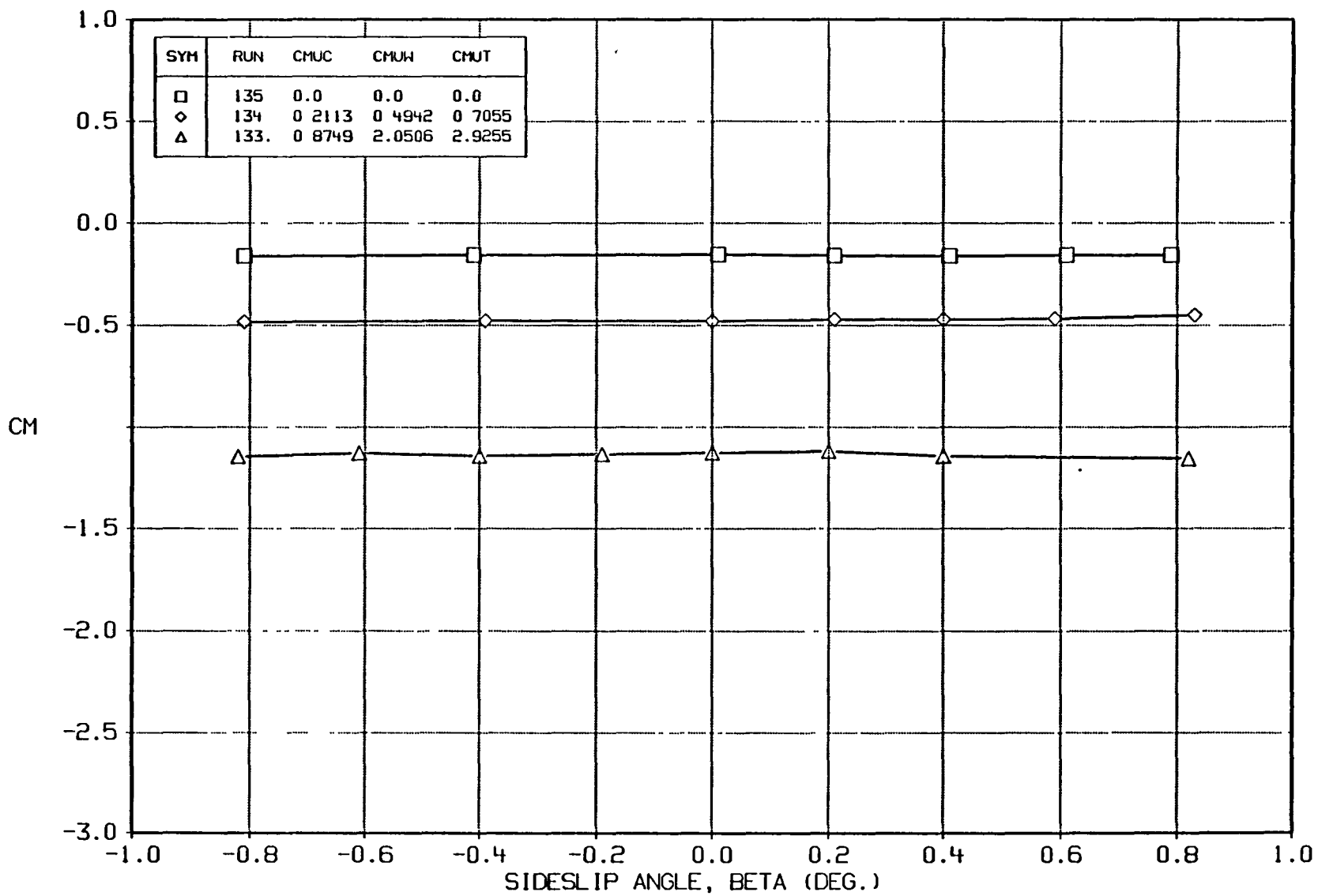


FIGURE A25c BASIC DATA EFFECT OF CMU  
CONFIGURATION BWGV, BN/B=1.0, DELF=45,  $\alpha = 2$ .

A-156

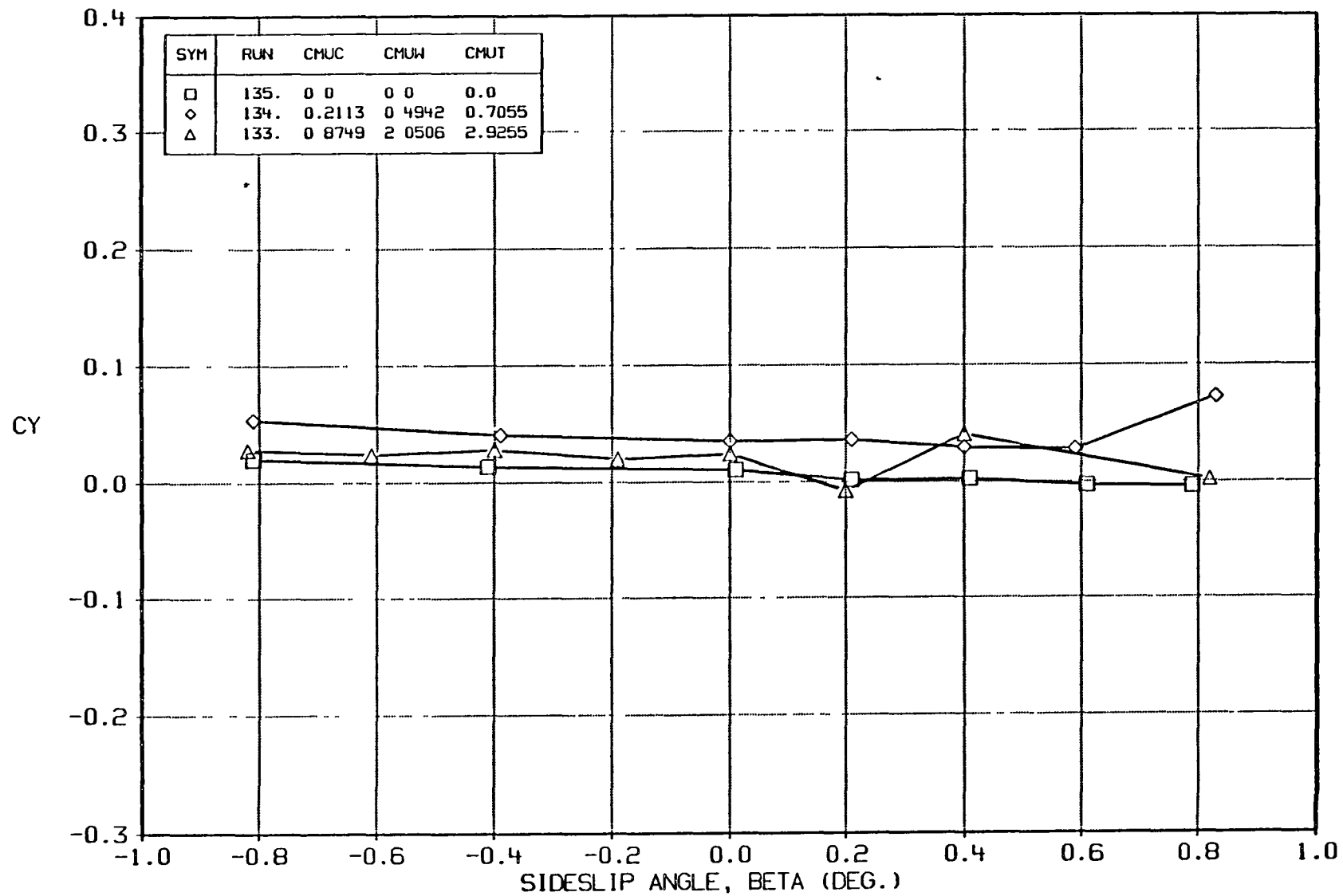


FIGURE A25d BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V,  $B_N/B=1.0$ ,  $\Delta E_L=45$ ,  $\alpha = 2$

A-157

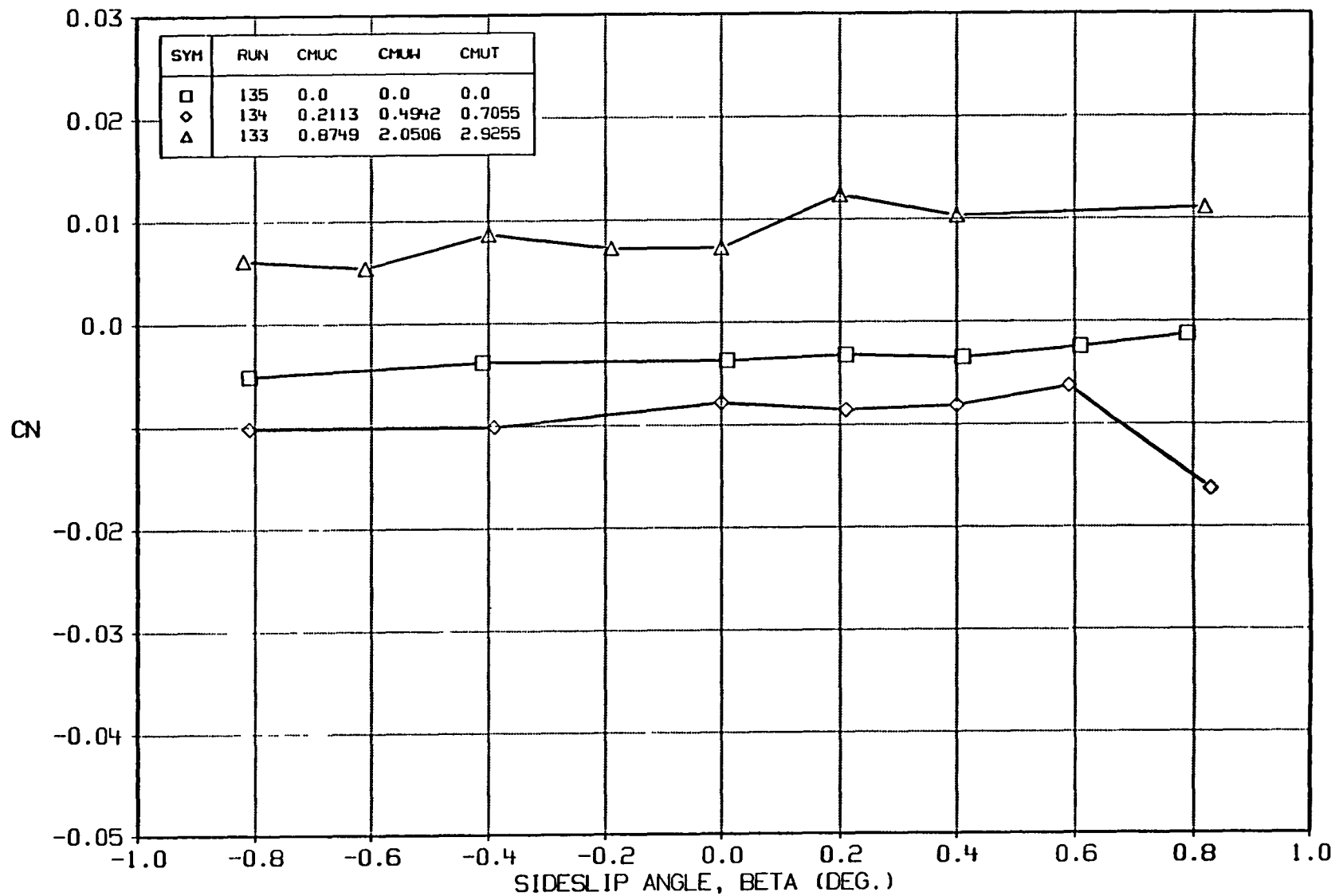


FIGURE A25e BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=1.0, DELF=45,  $\alpha = 2$

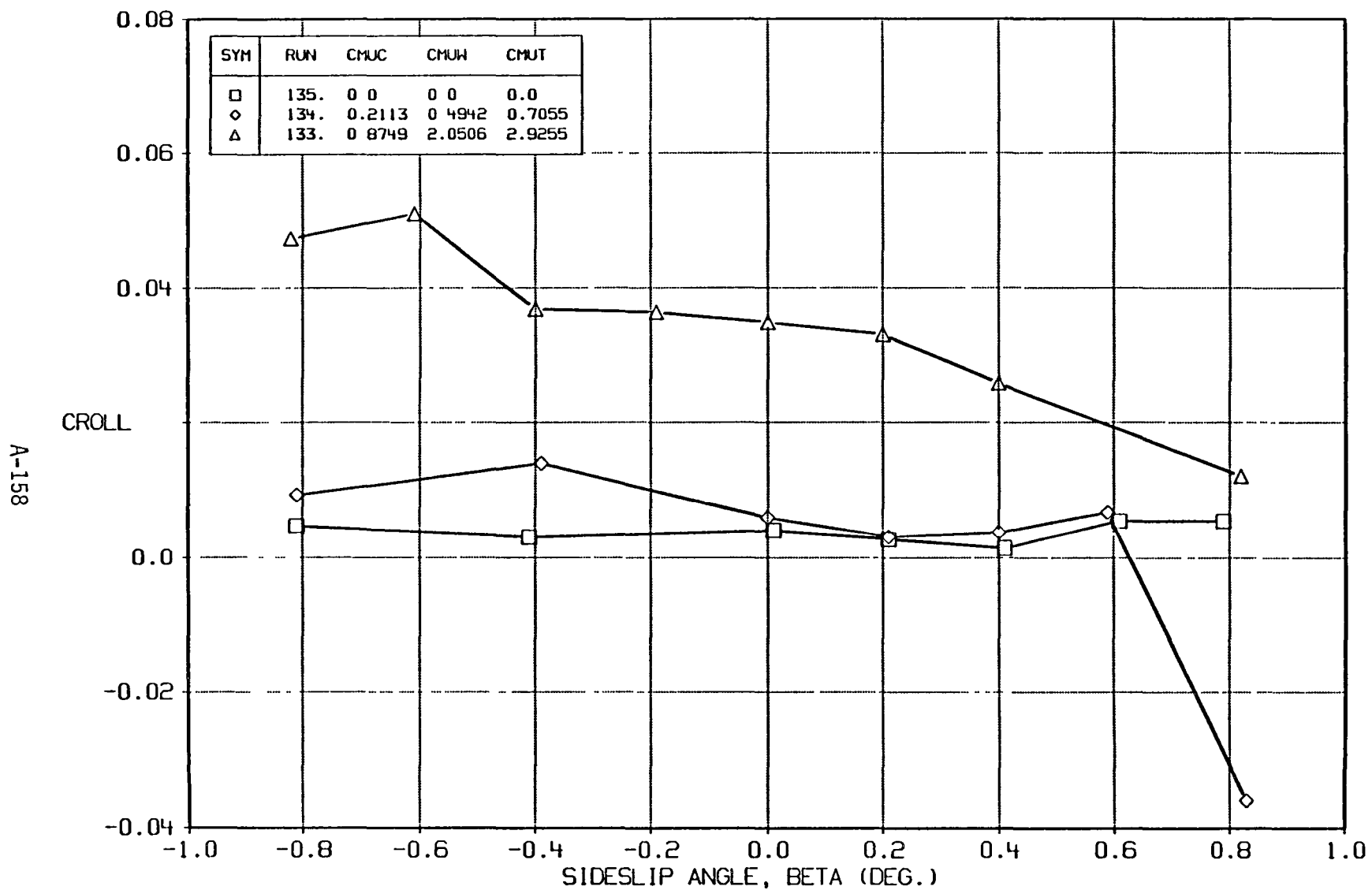


FIGURE A25f BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V, BN/B=1.0, DELF=45,  $\alpha = 2$



A-159

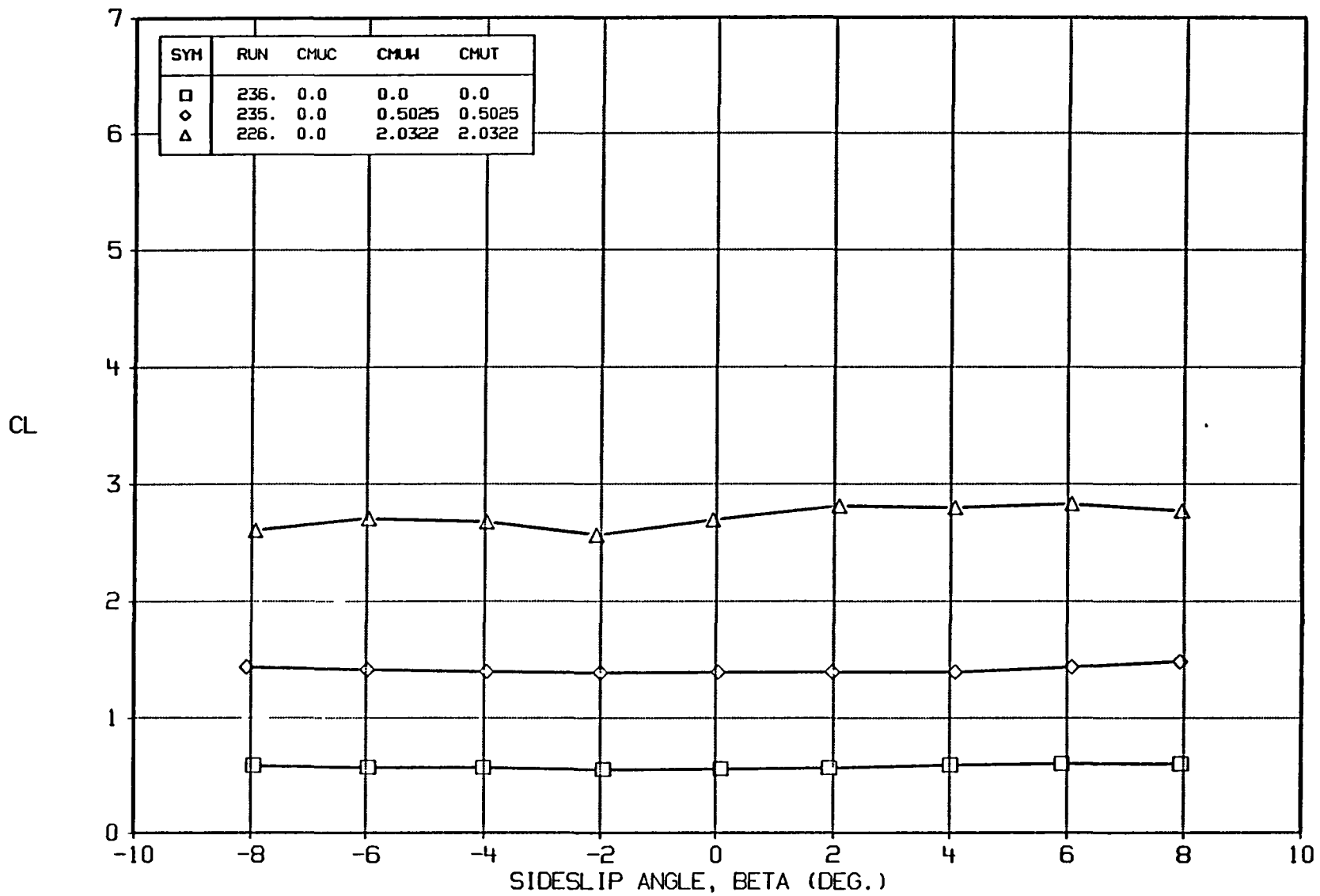


FIGURE A26a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5 DELF=45

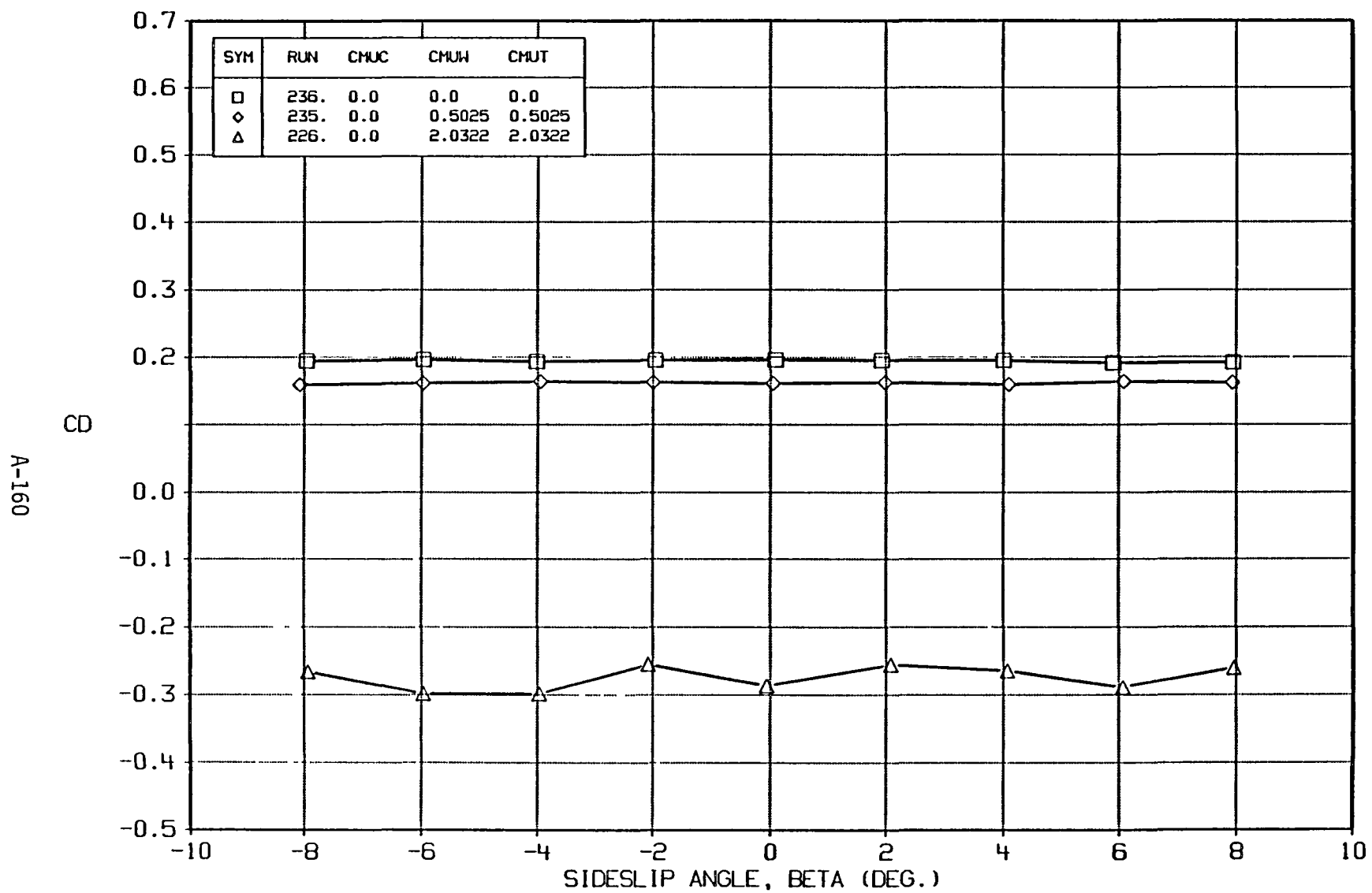


FIGURE A26b BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V, BN/B=0.5 DELF=45

A-161

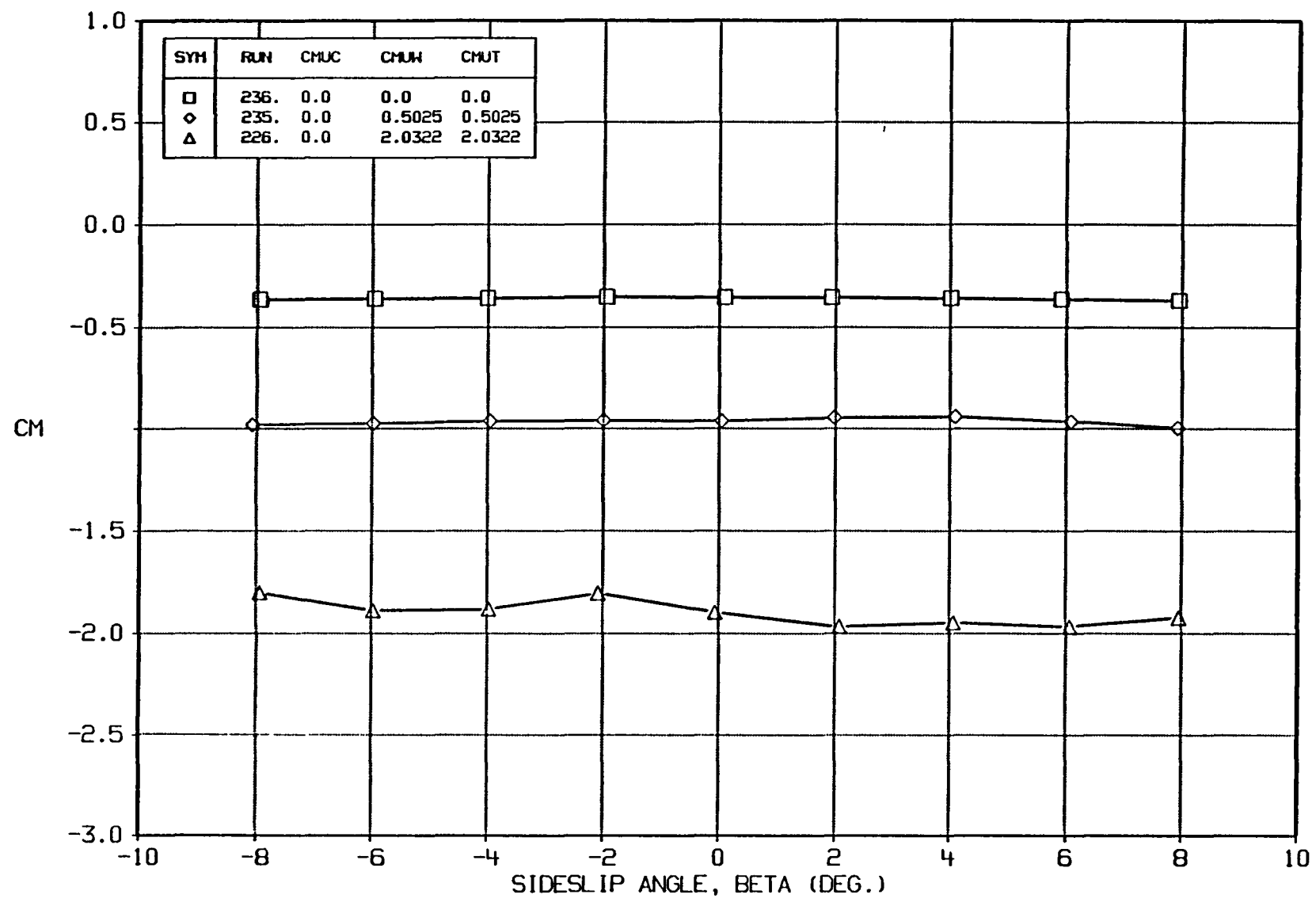


FIGURE A26c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5 DELF=45

A-162

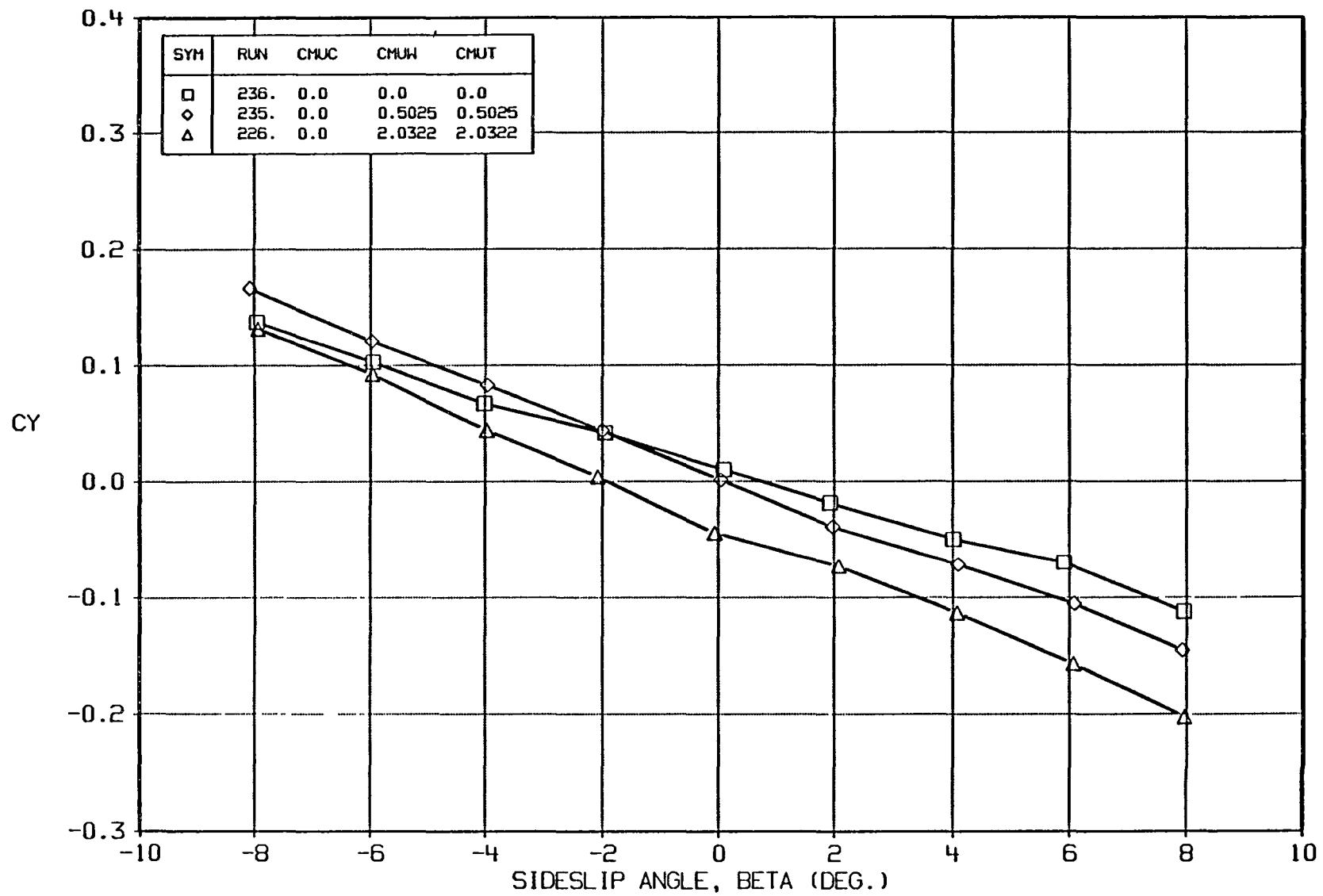


FIGURE A26d BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5 DELF=45

A-163

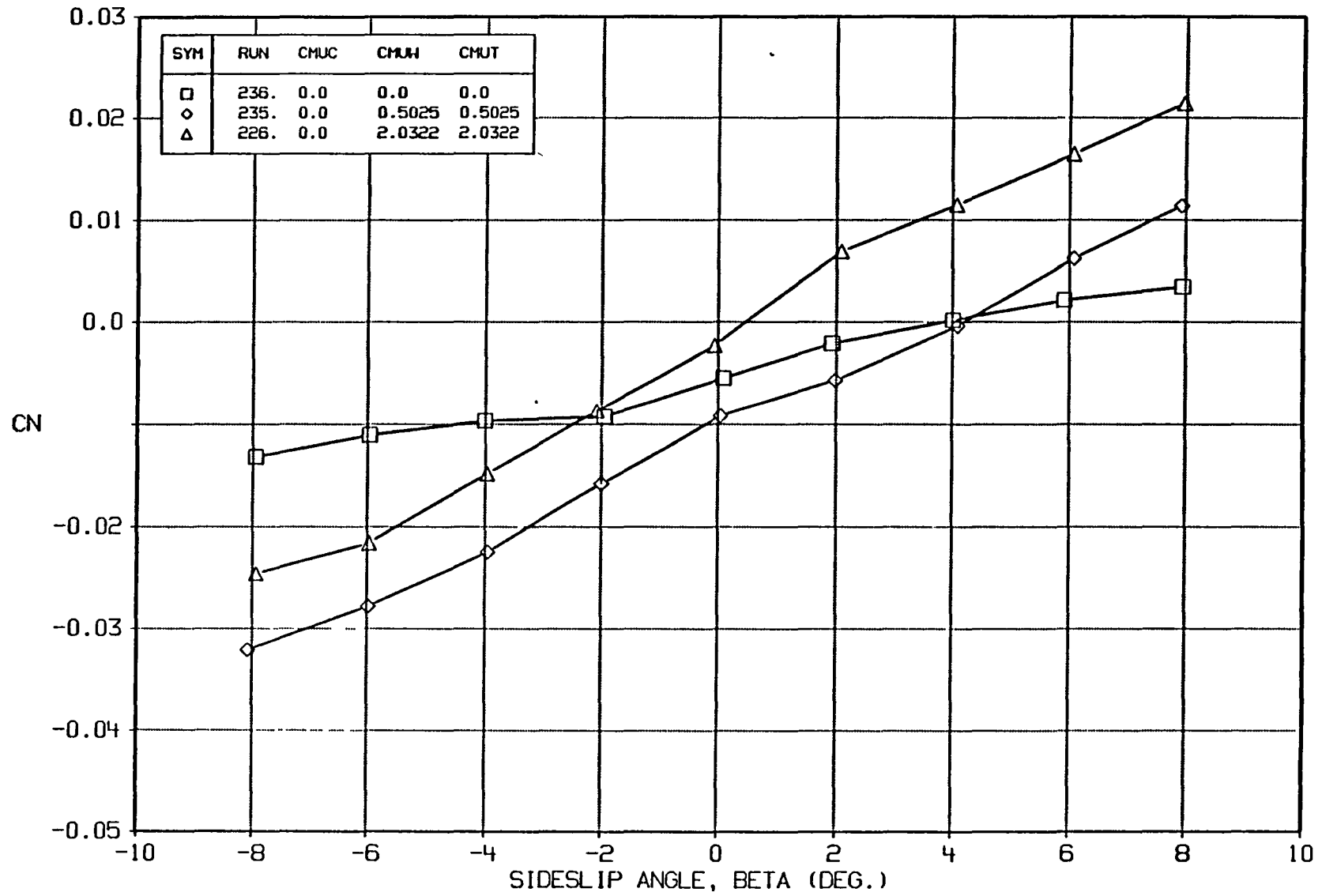


FIGURE A26e BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5 DELF=45

A-164

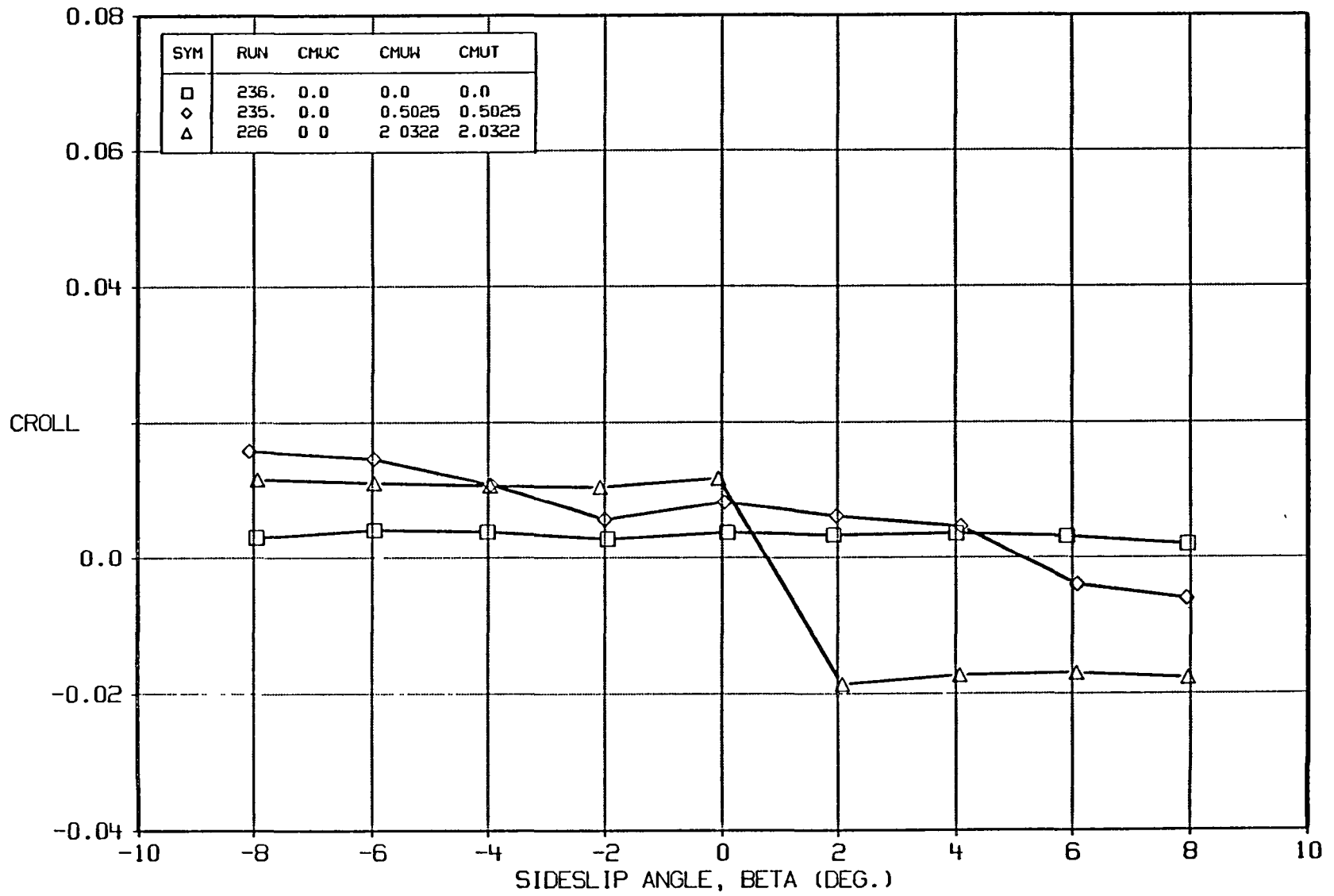


FIGURE A26f BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, BN/B=0.5 DELF=45

A-165

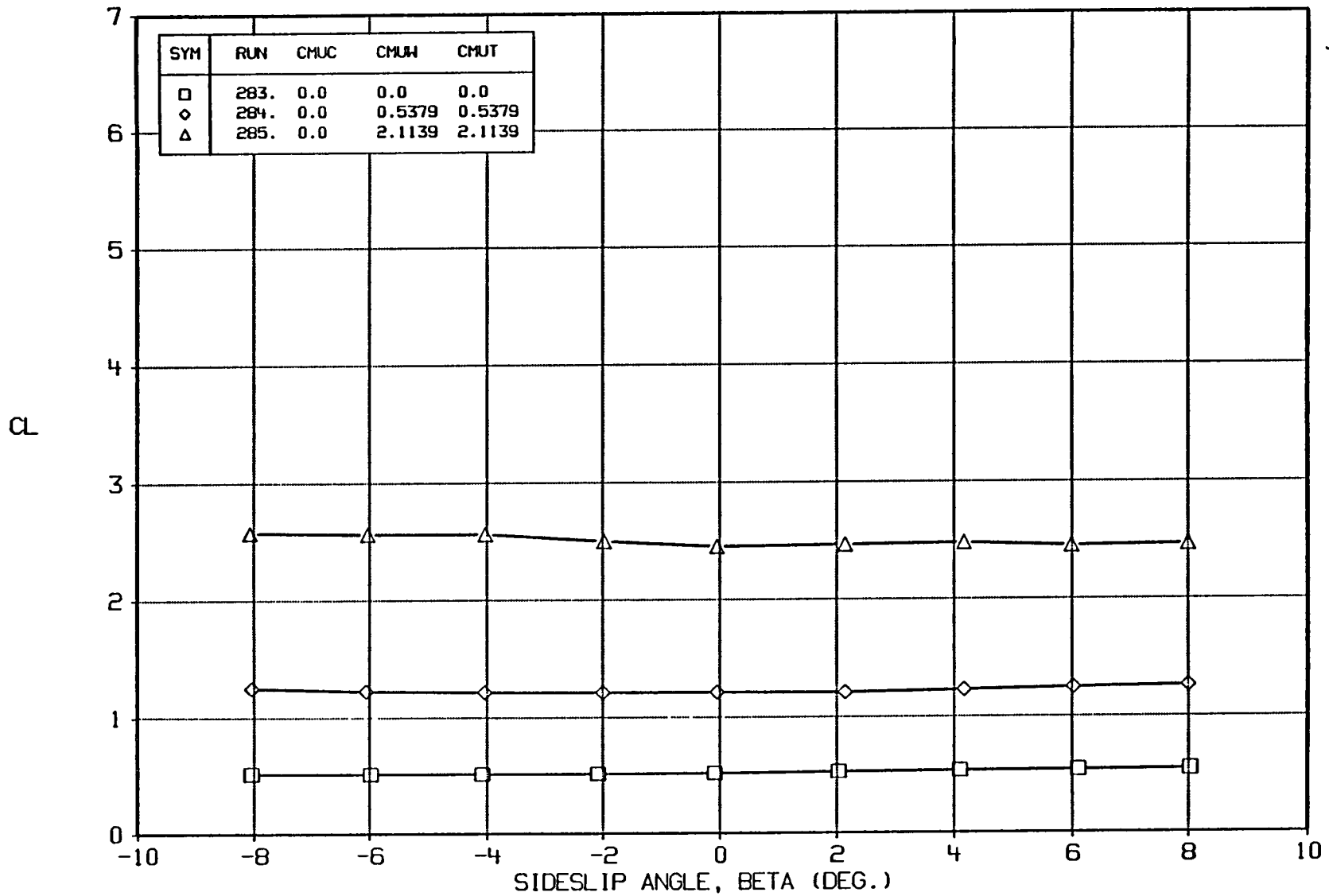


FIGURE A27a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25

A-166

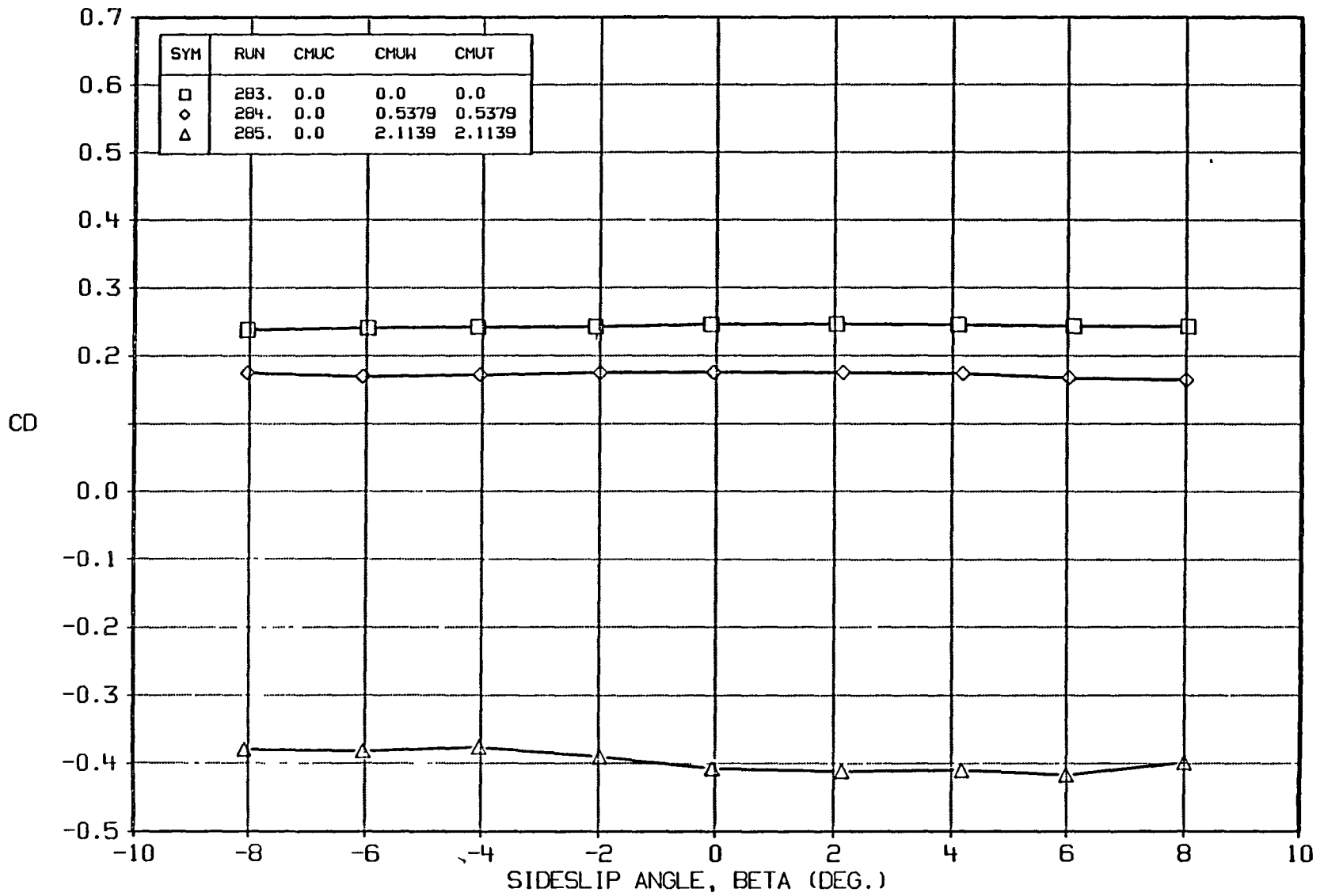


FIGURE A27b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25



A-167

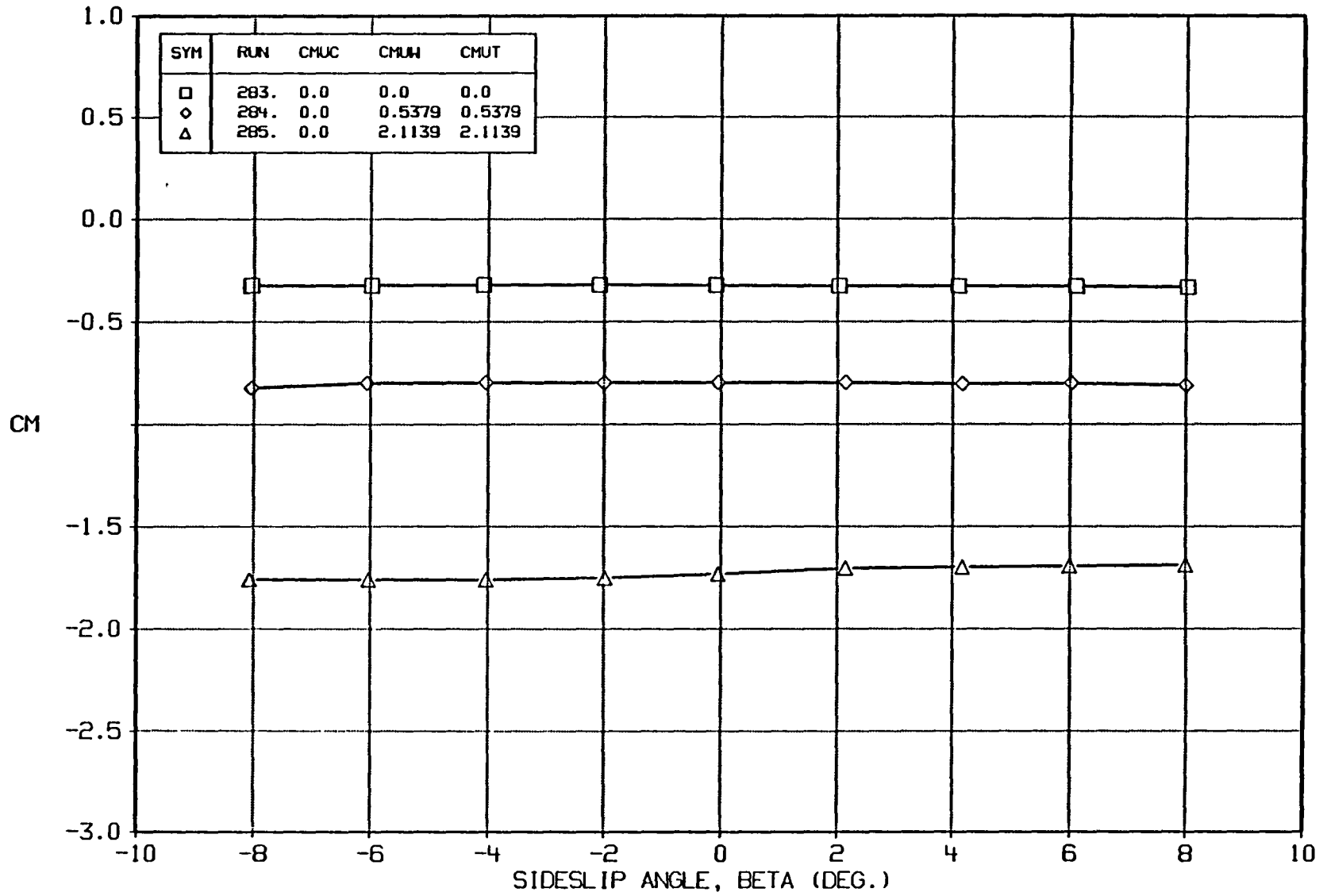


FIGURE A27c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25

A-168

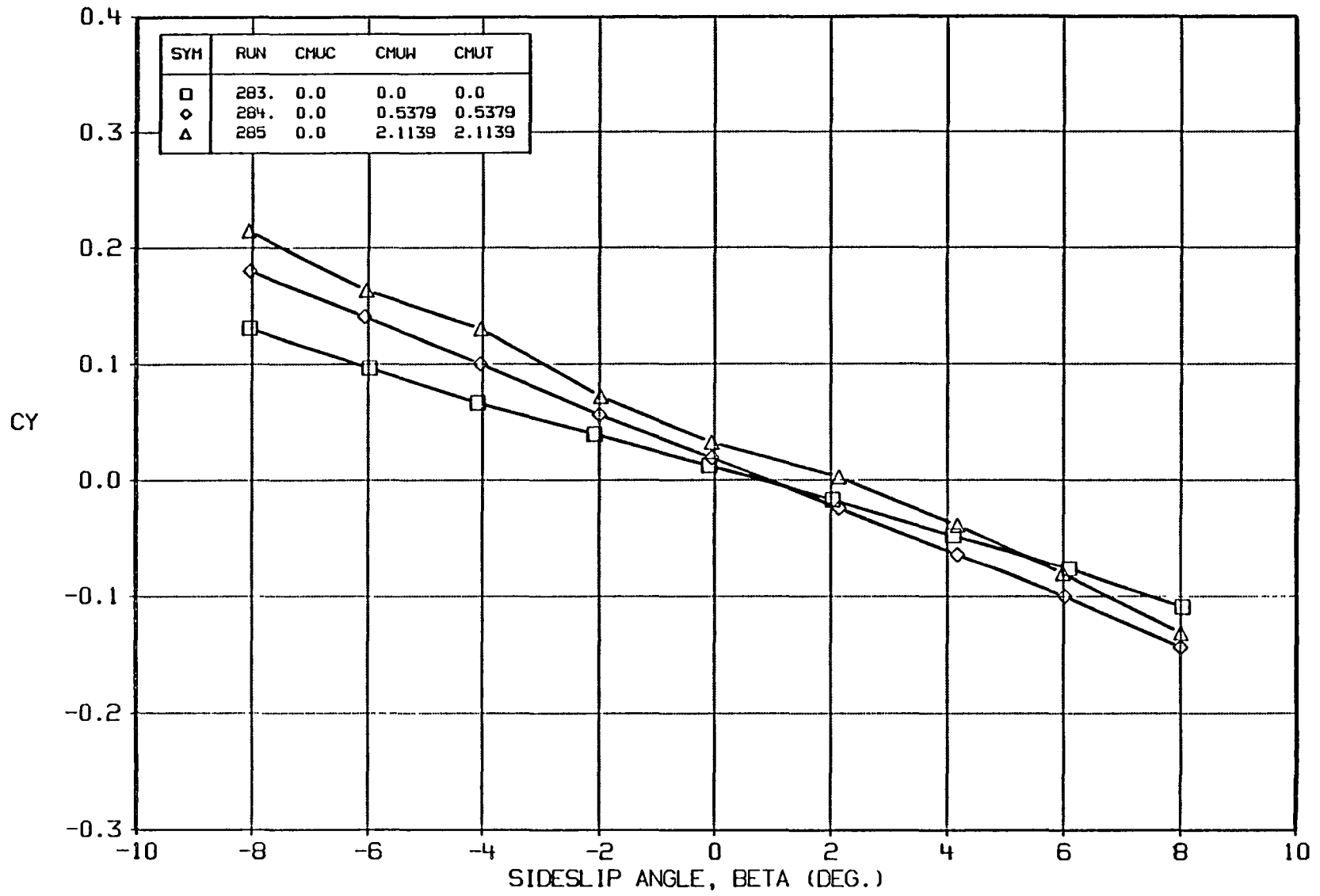


FIGURE A27d BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25

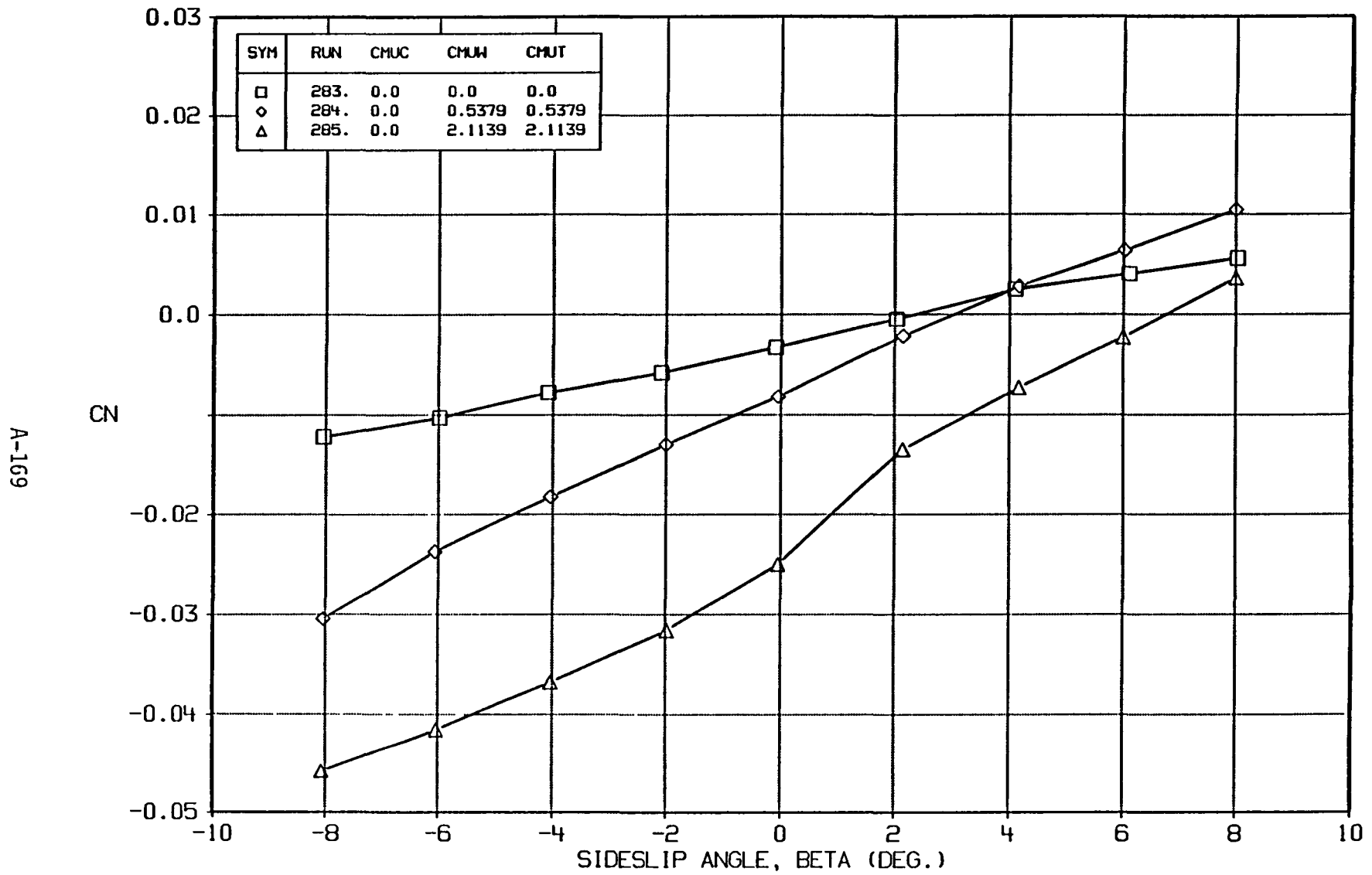


FIGURE A27e BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V, DELF=45, BN/B=.25

A-170

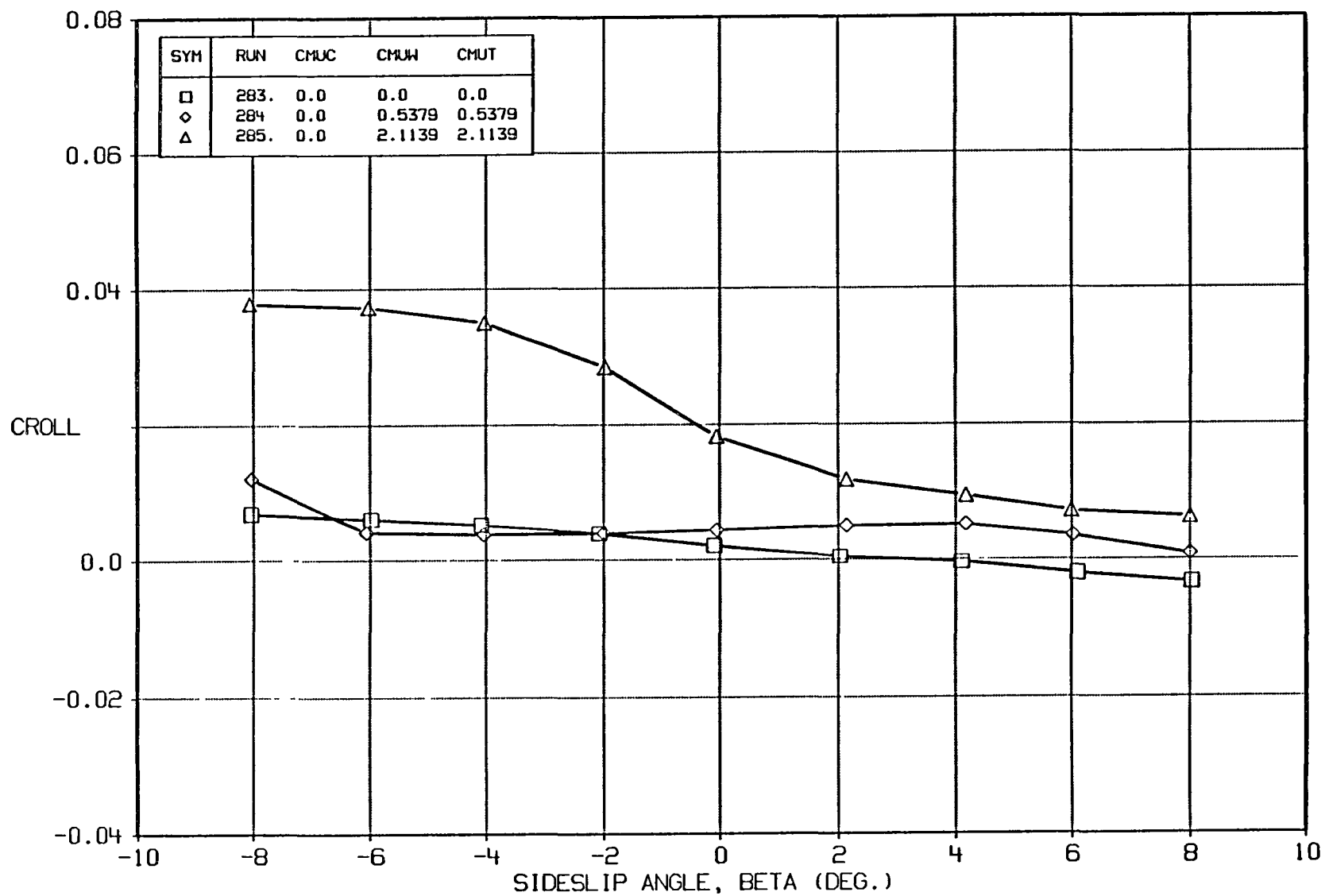


FIGURE A27f BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=.25

A-171

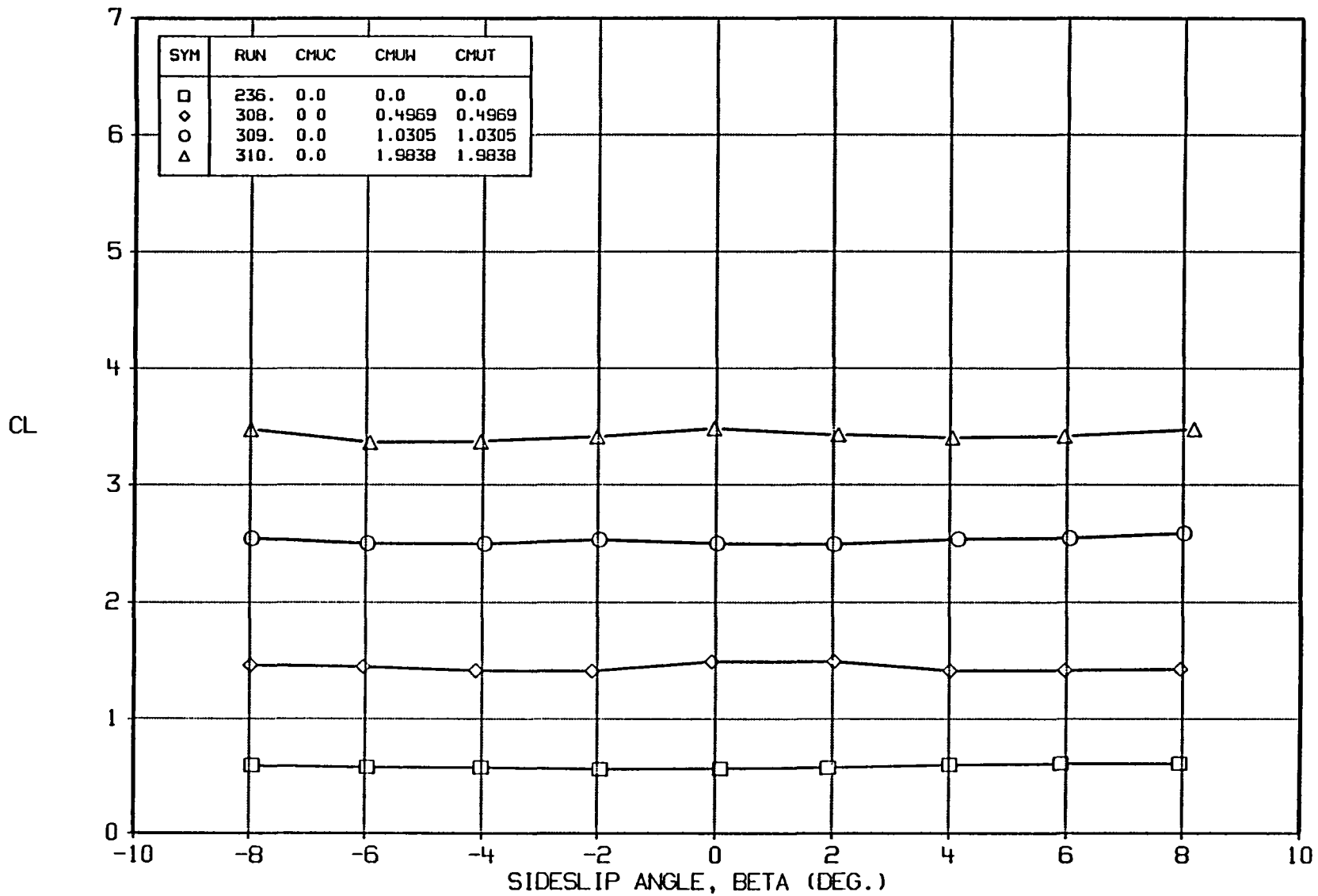


FIGURE A28a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

A-172

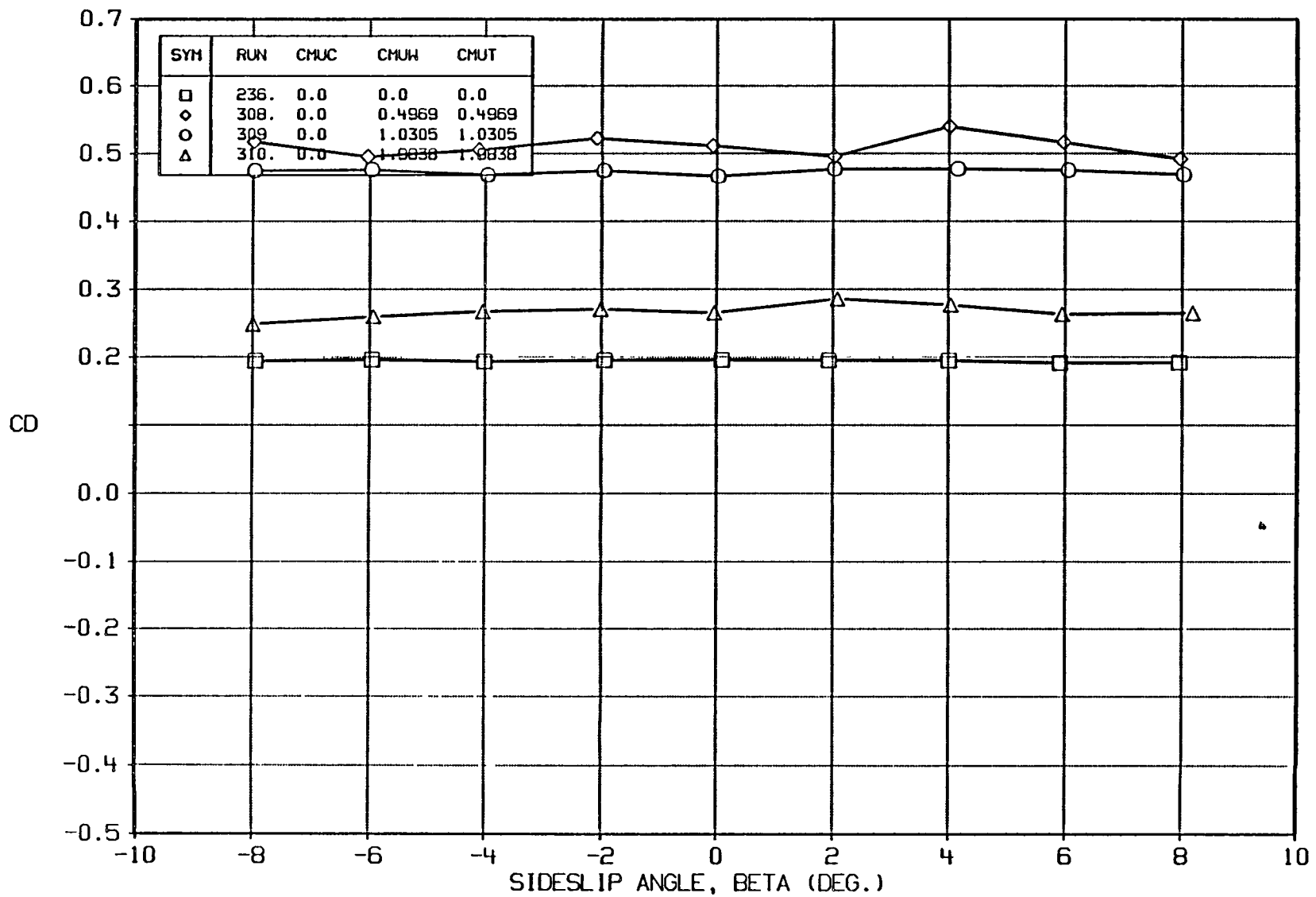


FIGURE A28b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

A-173

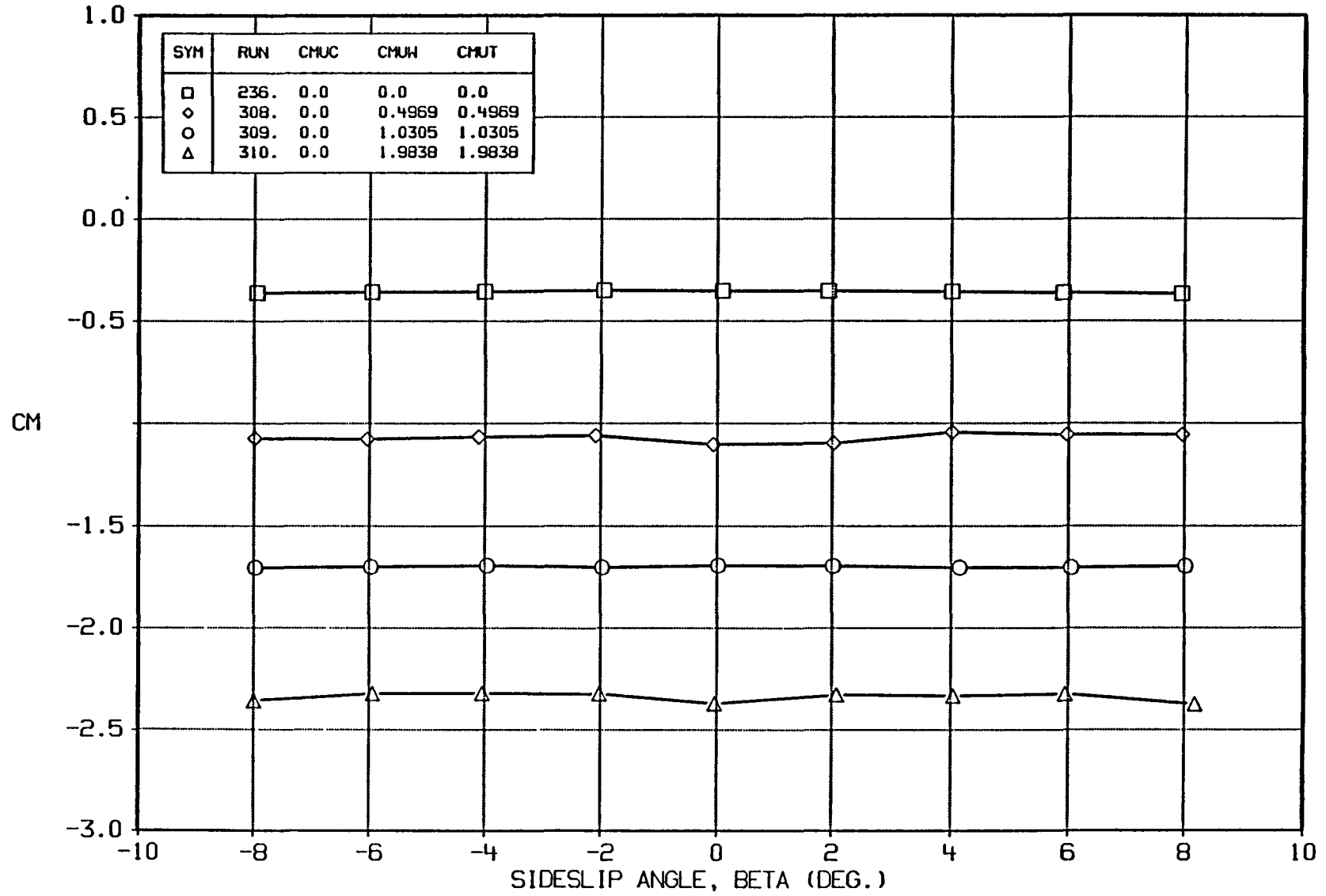


FIGURE A28c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

A-174

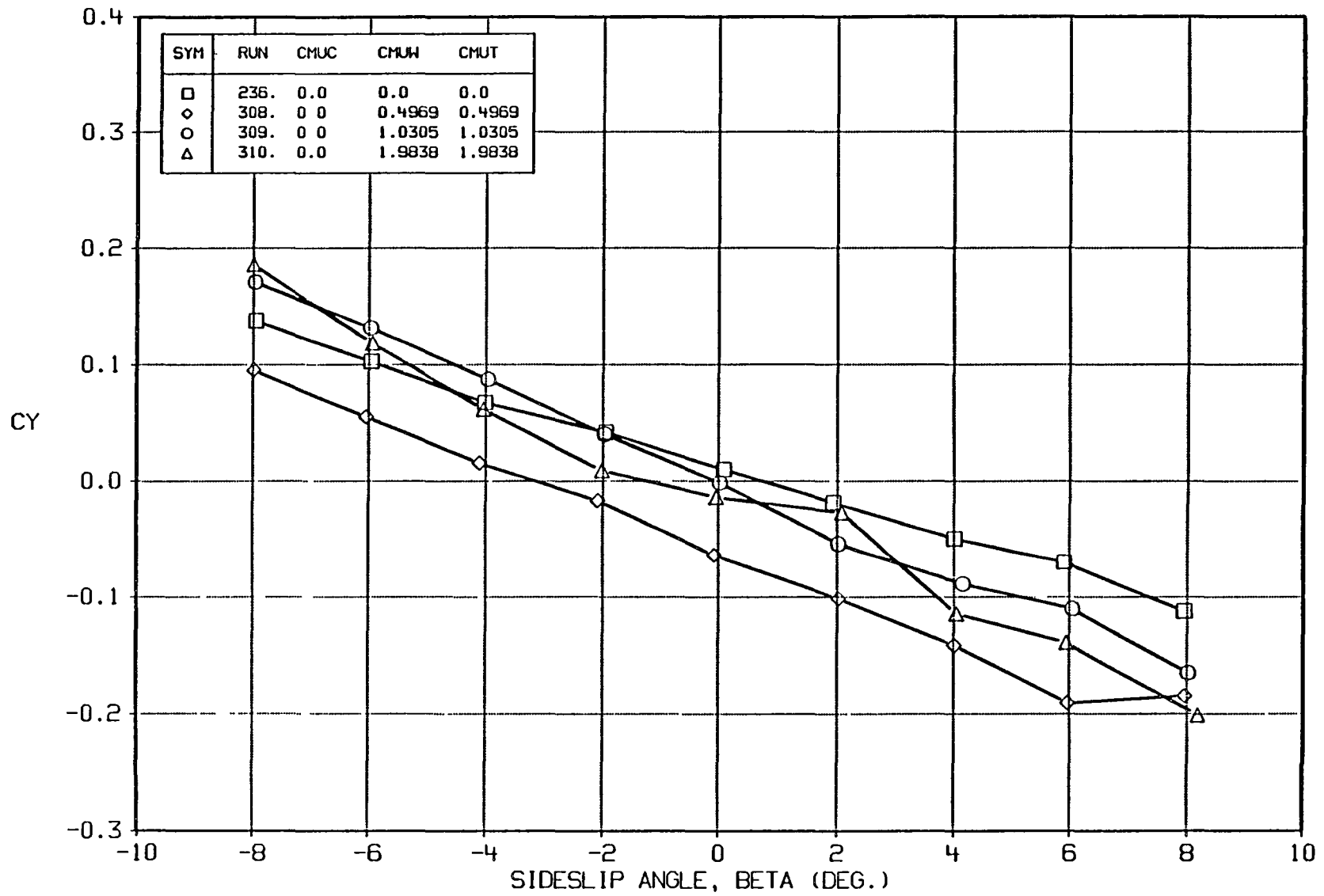


FIGURE A28d BASIC DATA EFFECT OF CMU  
CONFIGURATION BWSV, DELF=45, BN/B=1



A-175

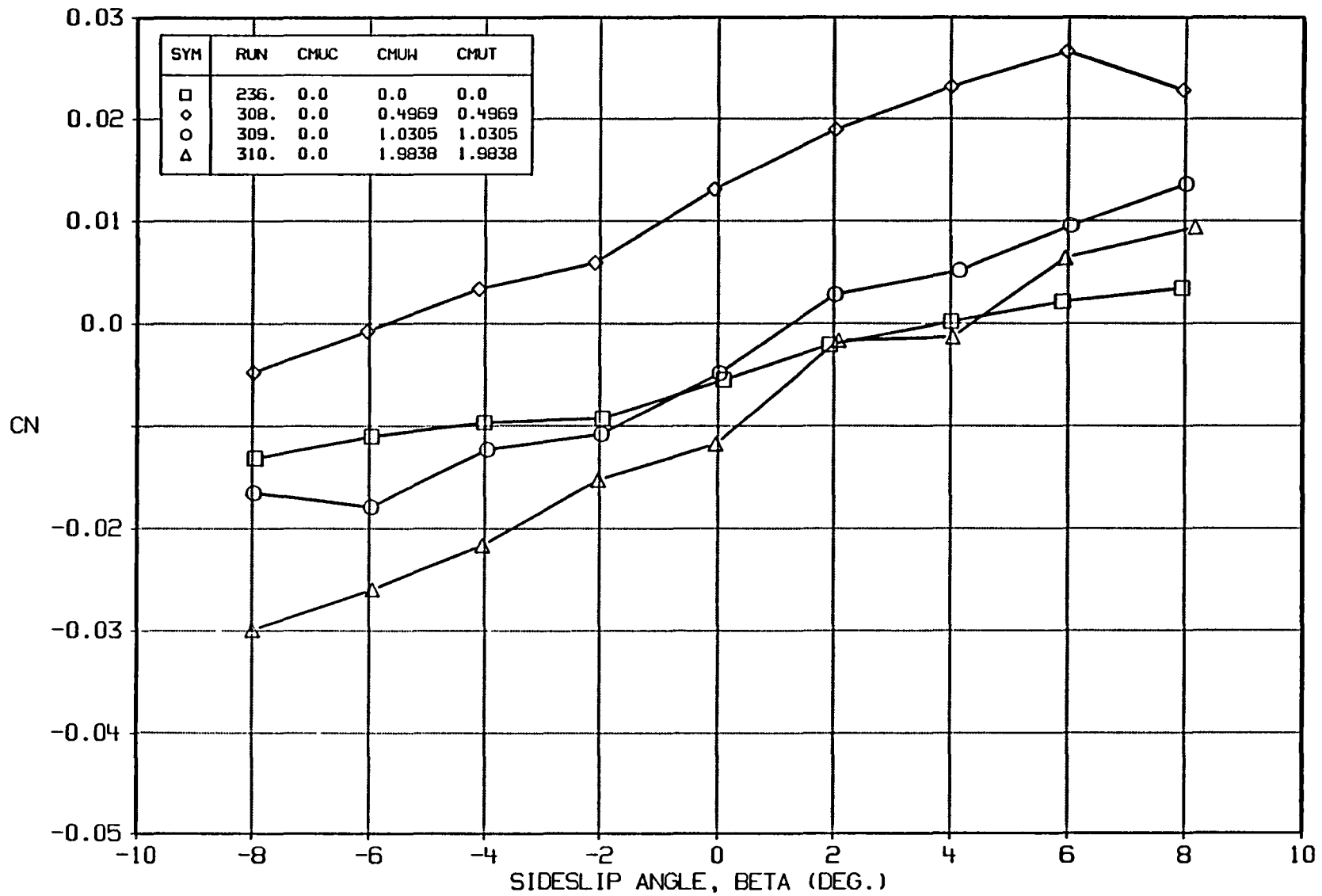


FIGURE A28e BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

A-176

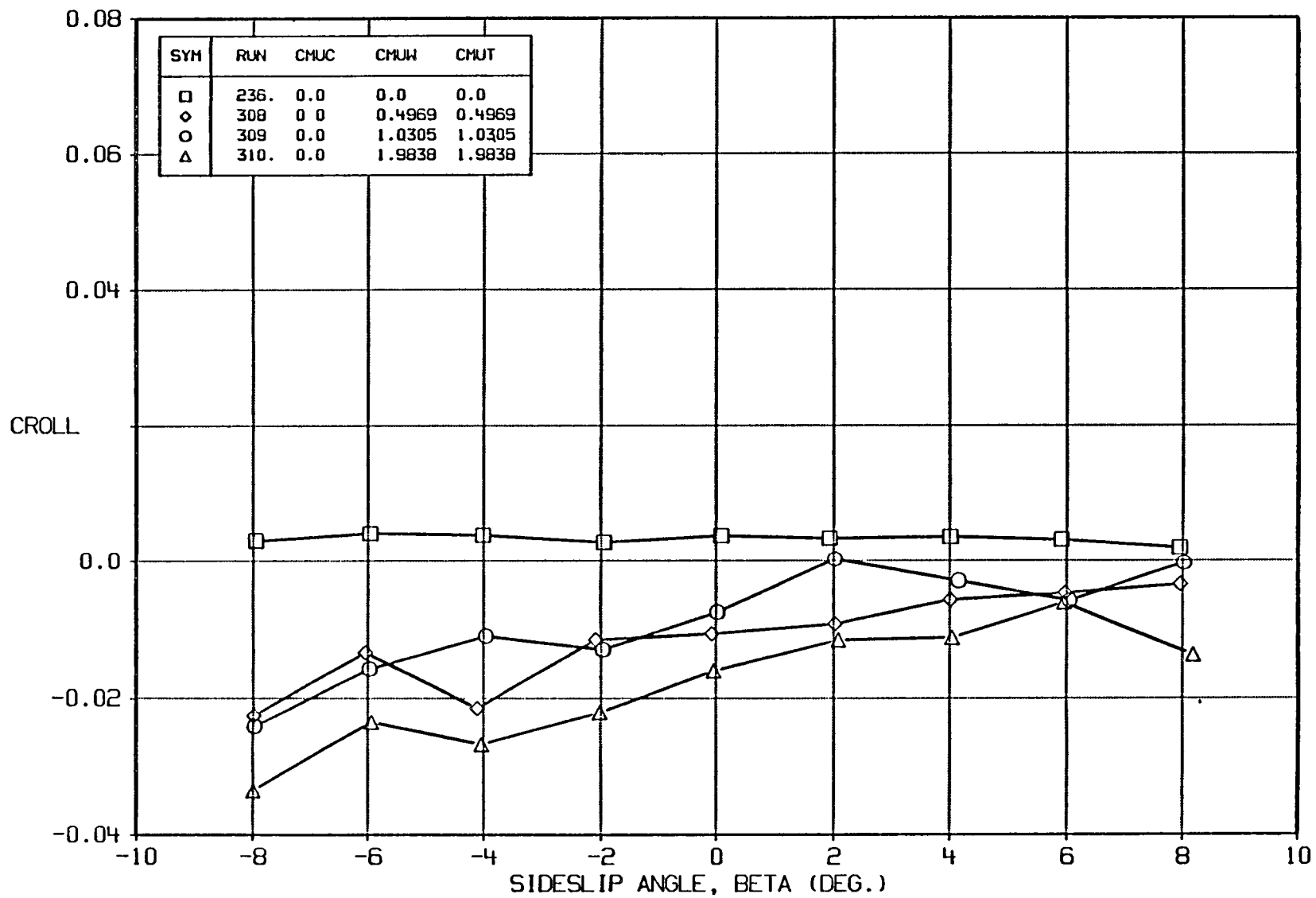


FIGURE A28f BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, BN/B=1

A-177

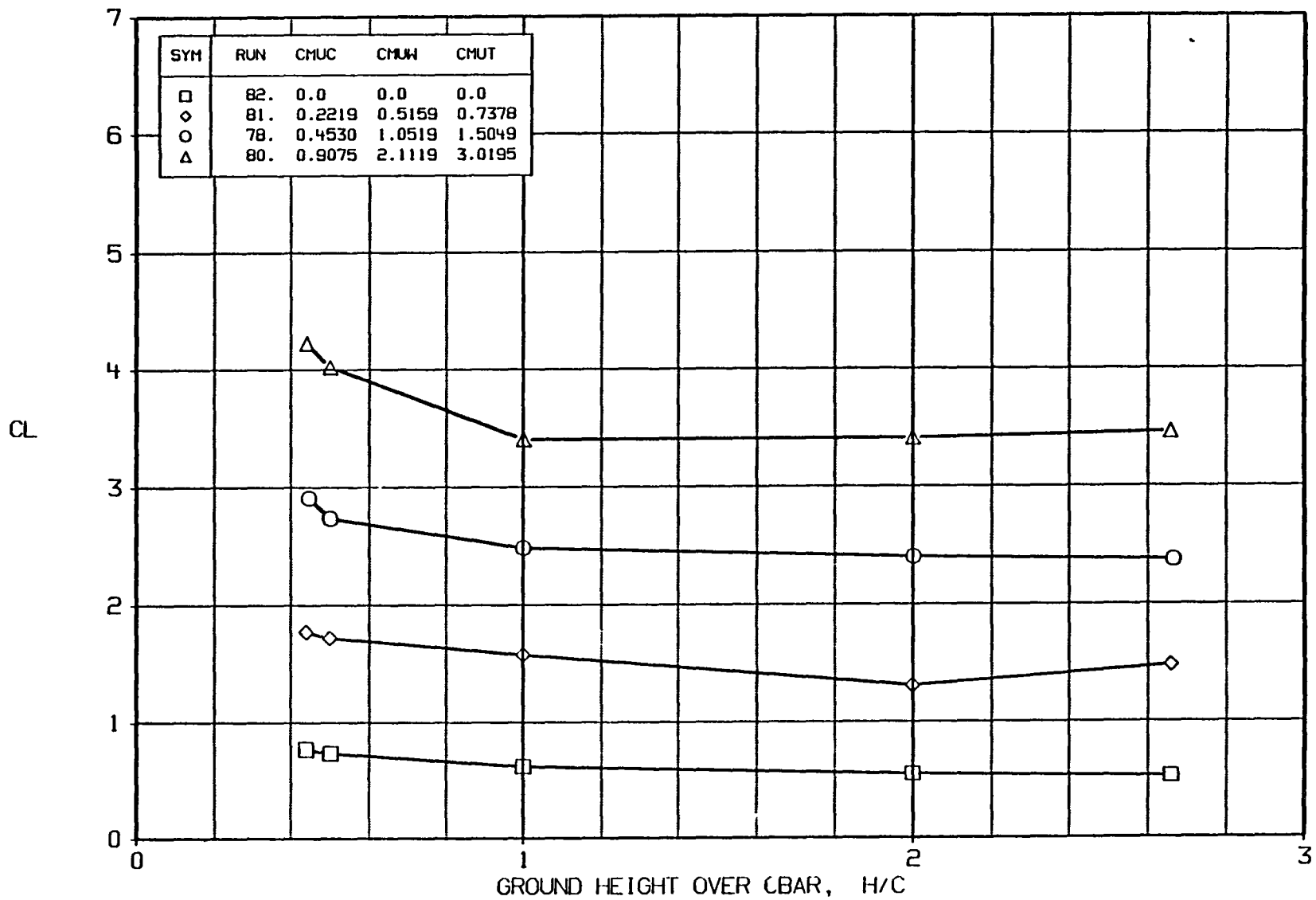


FIGURE A29a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-178

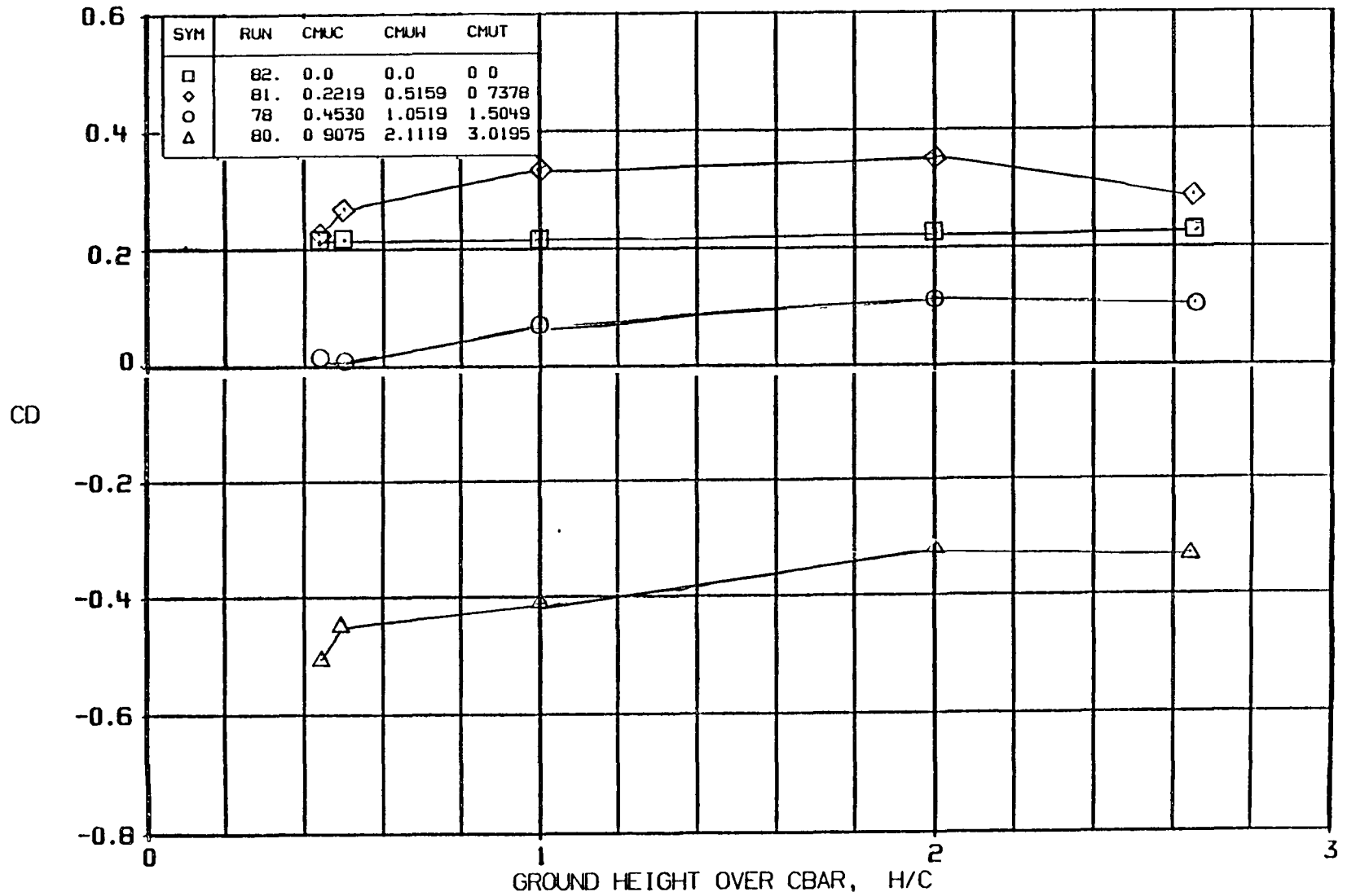


FIGURE A29b BASIC DATA EFFECT OF CMU CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-179

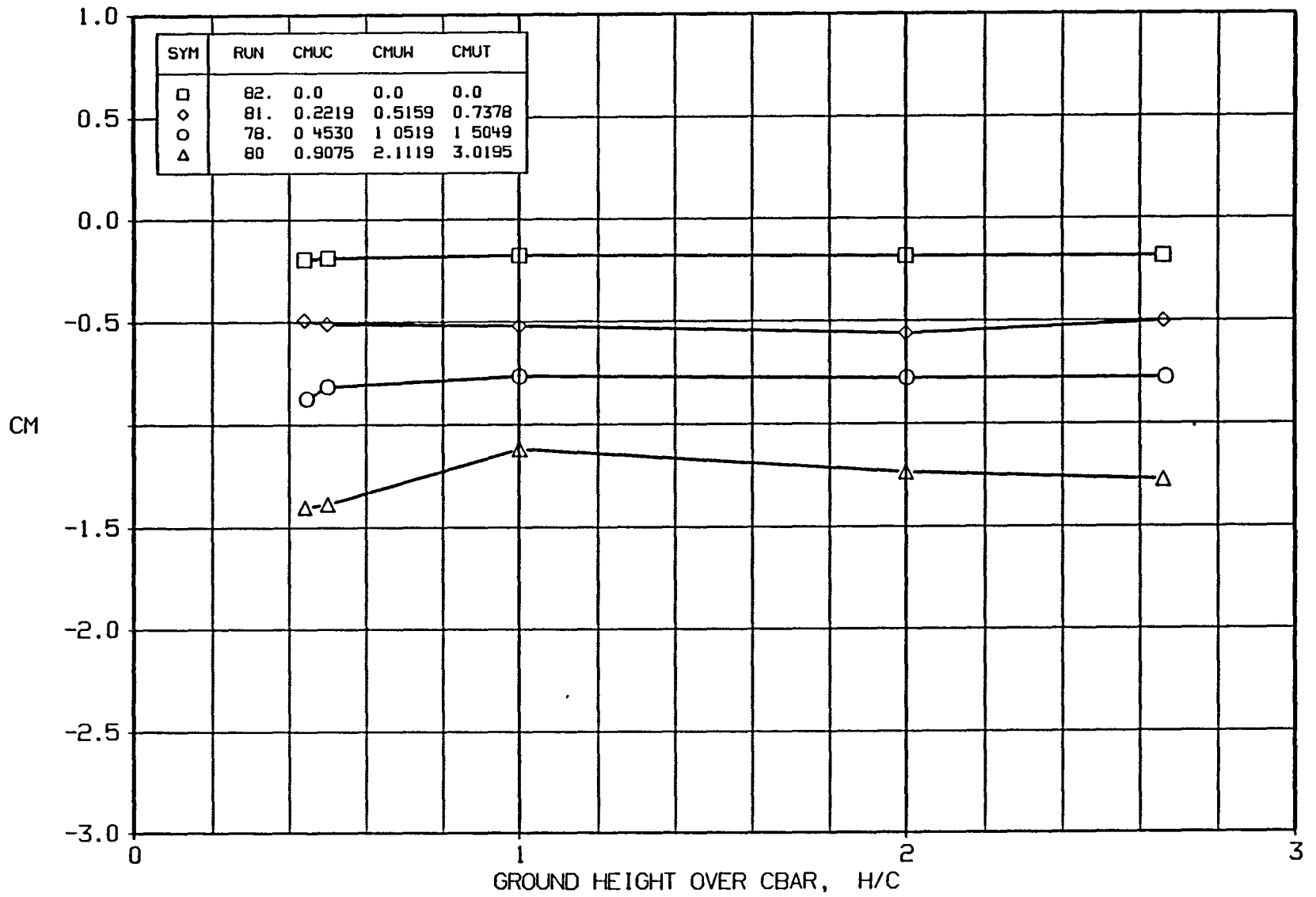


FIGURE A29c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1

A-180

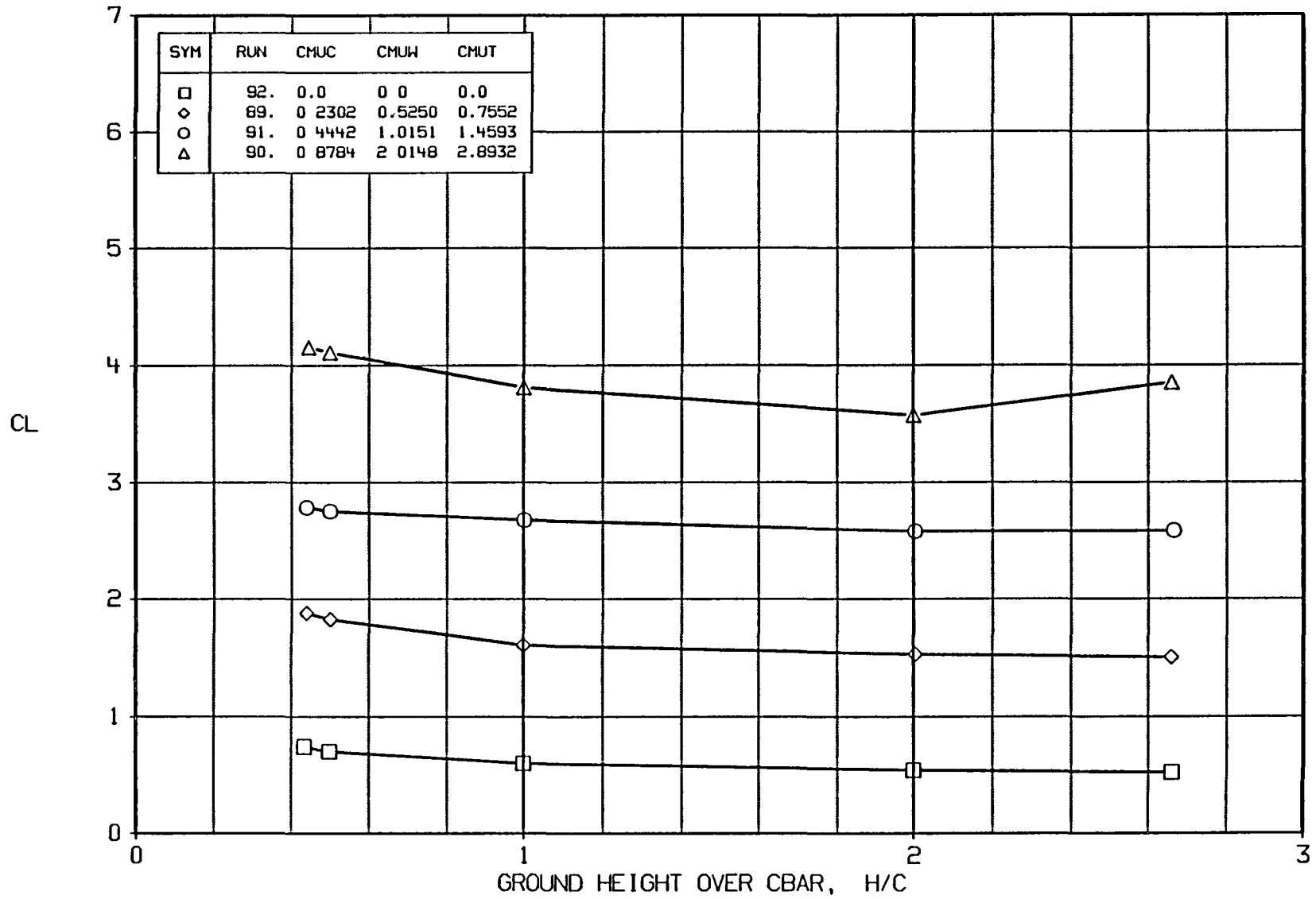


FIGURE A30a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W6V, DELF=45, BN/B=1

A-181

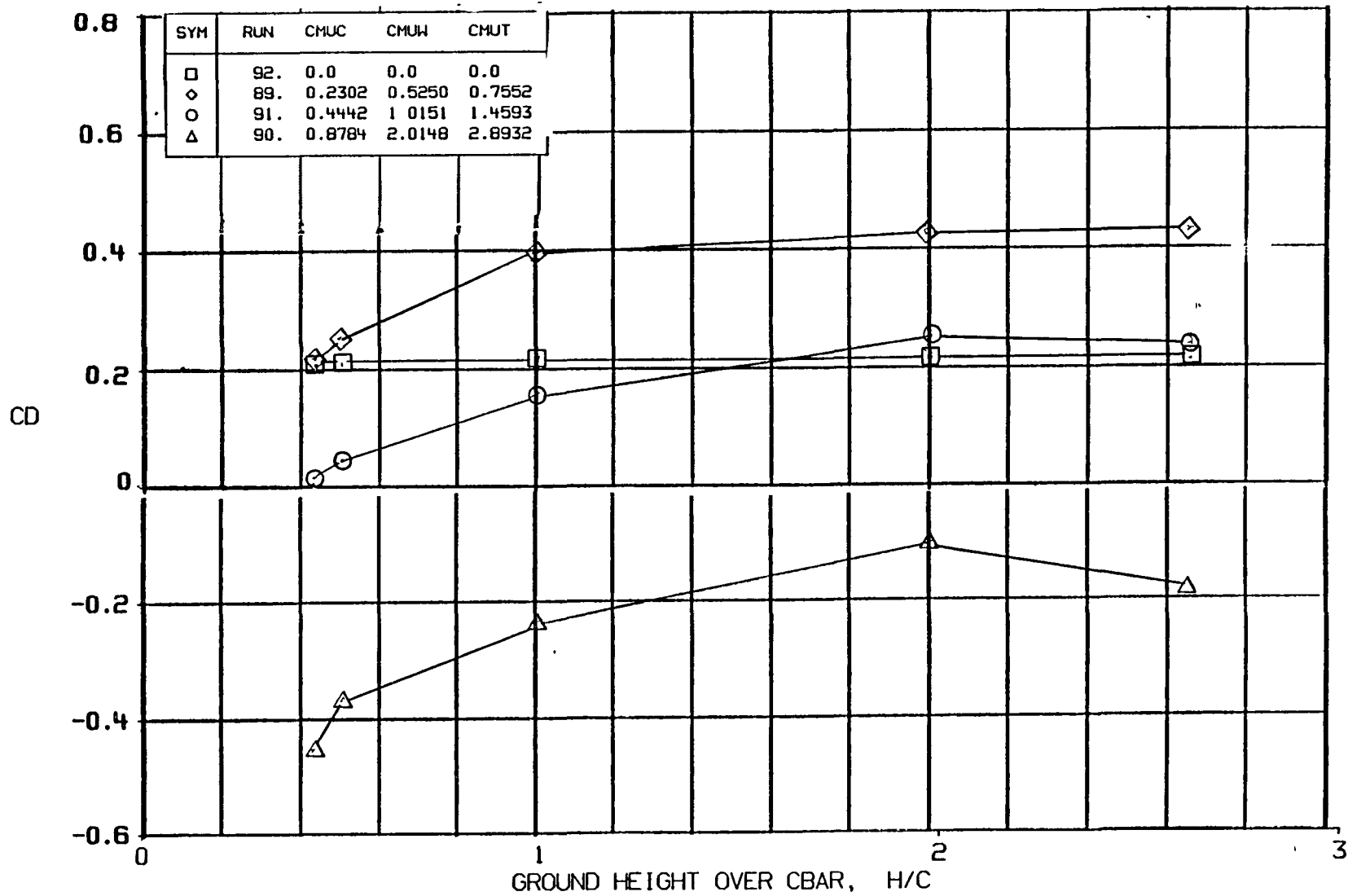


FIGURE A30b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W6V, DELF=45, BN/B=1

A-182

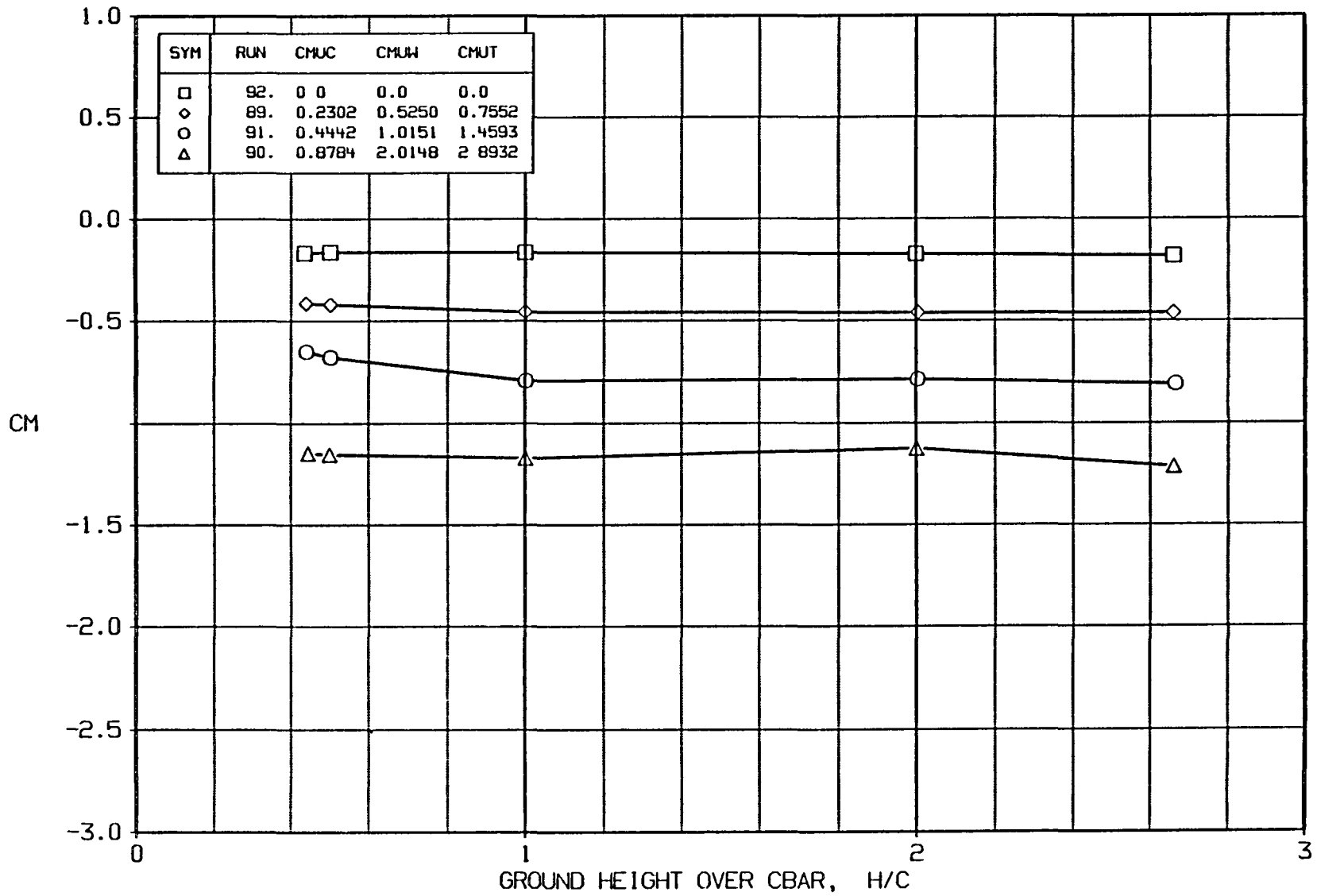


FIGURE A30c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W6V, DELF=45, BN/B=1.



A-183

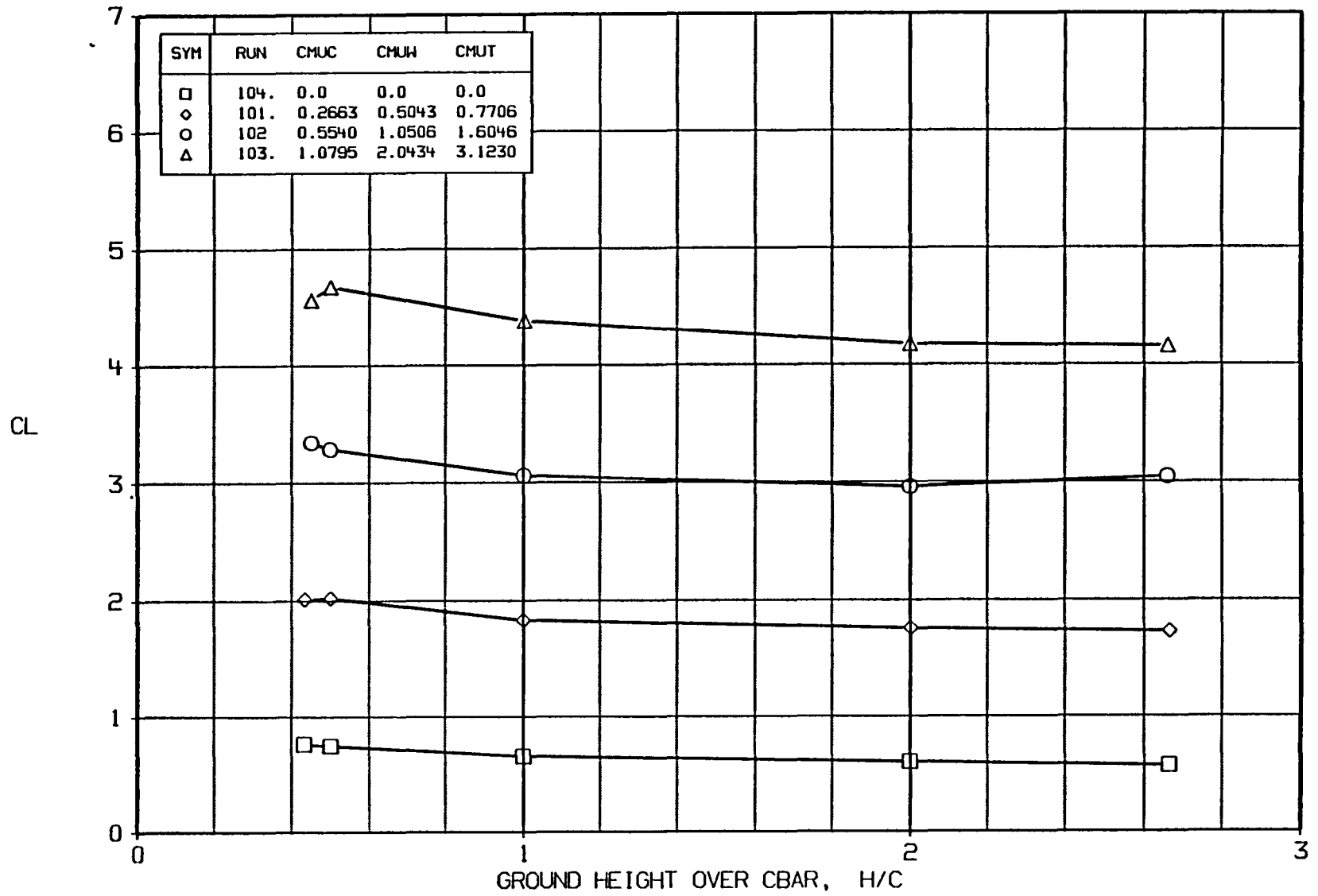


FIGURE A31a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W5V, DELF=45, BN/B=1

A-184

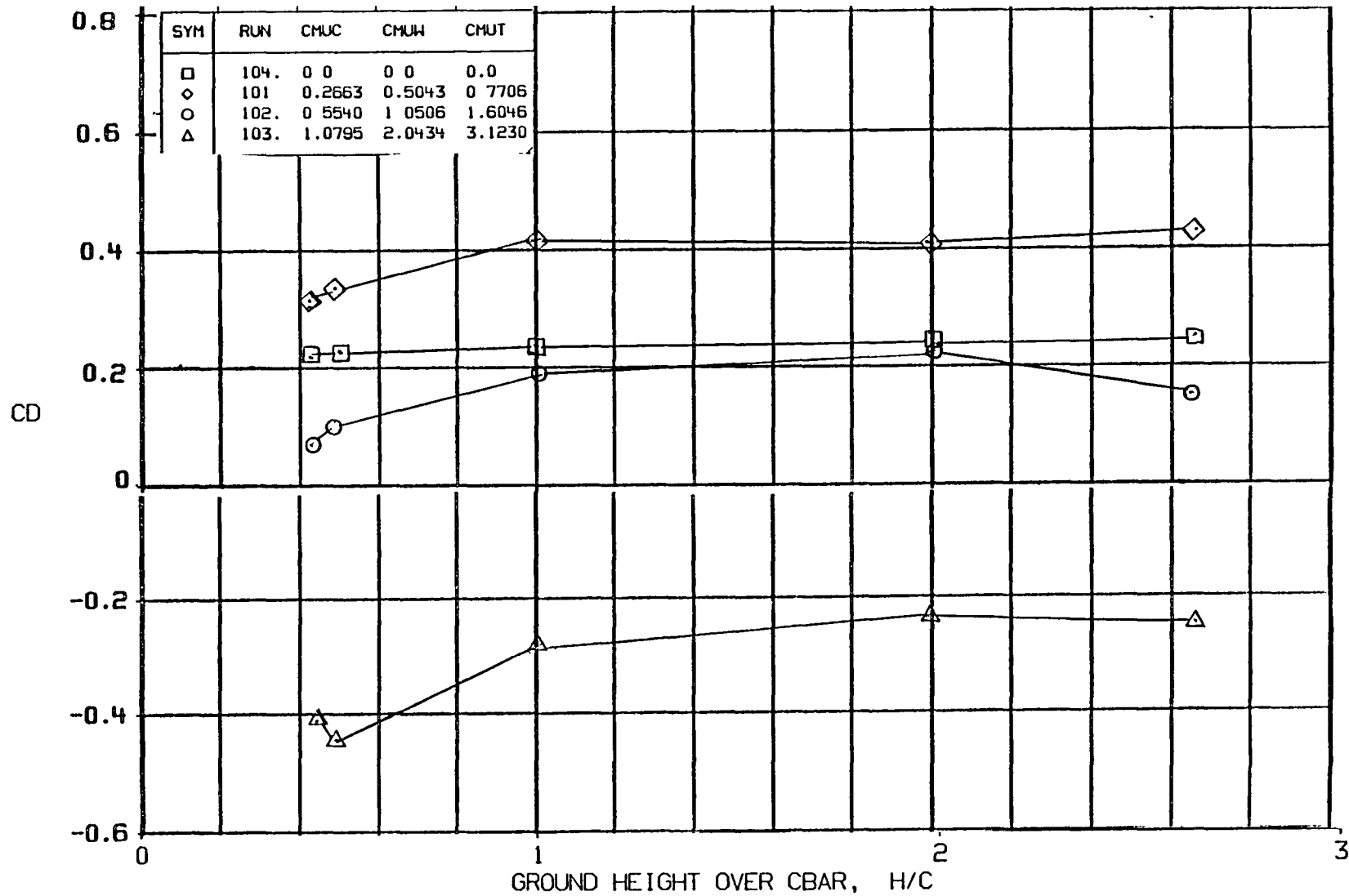


FIGURE A31b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W5V, DELF=45, BN/B=1

A-185

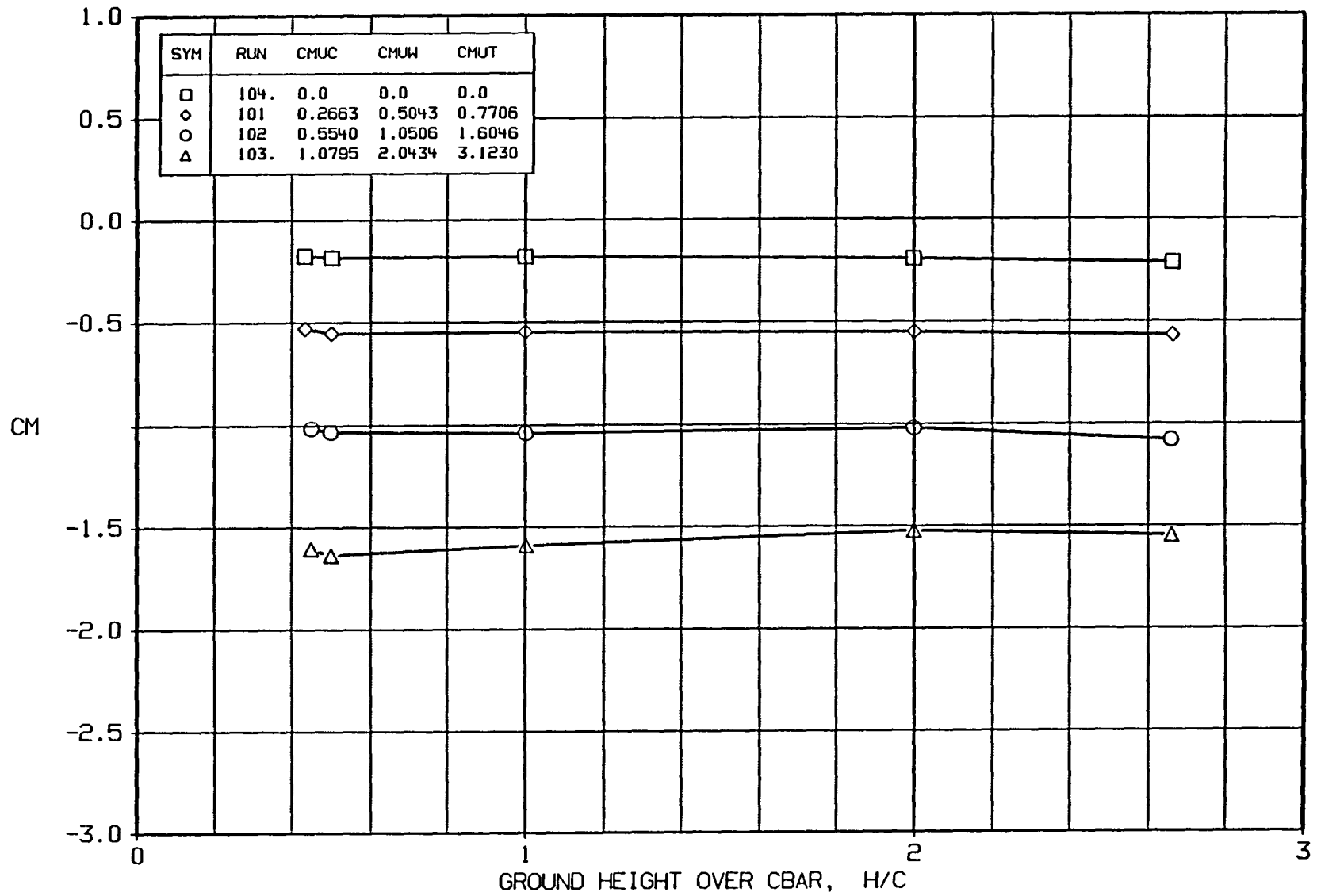


FIGURE A31c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC2W5V, DELF=45, BN/B=1

A-186

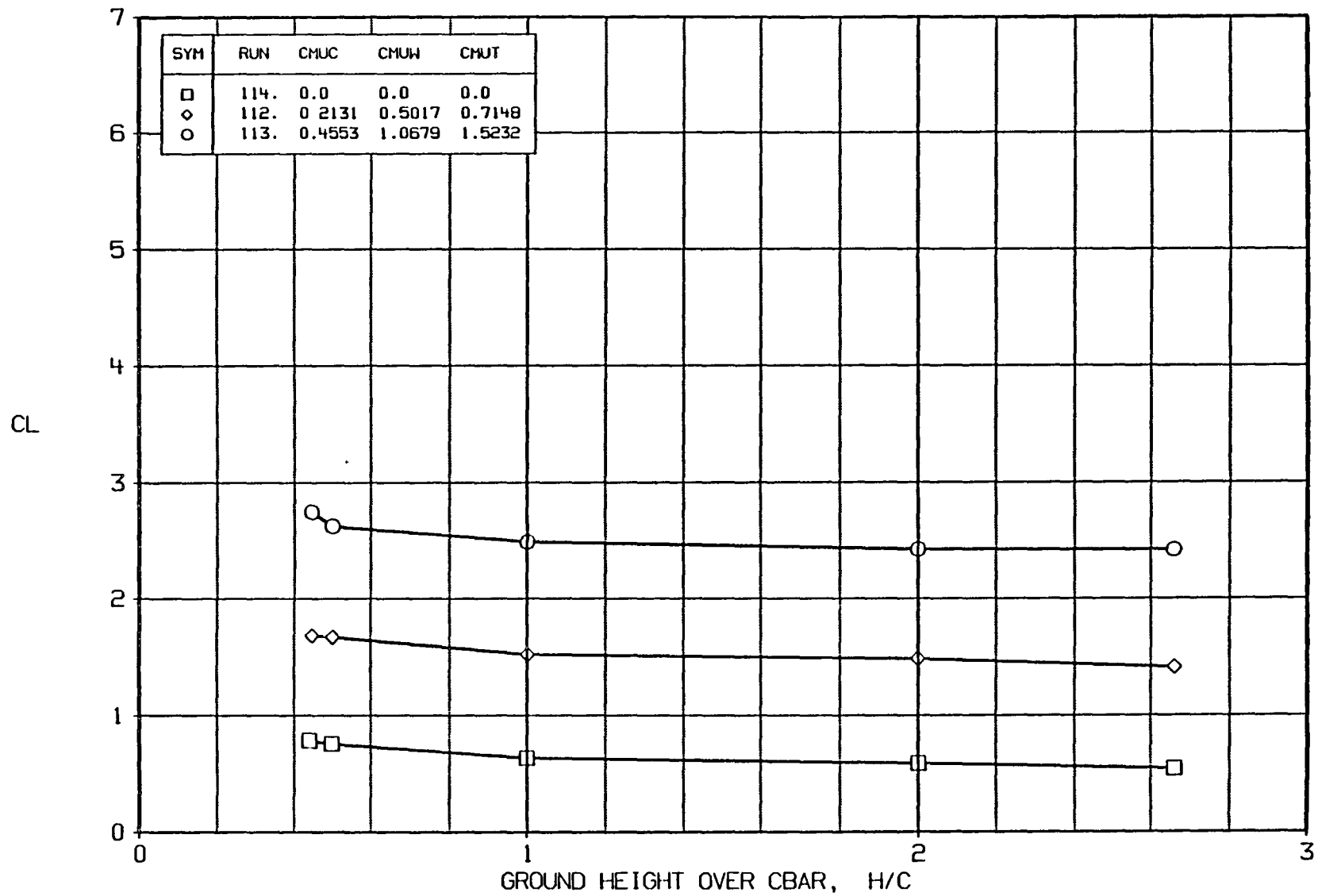


FIGURE A32a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC3W6V, DELF=45, BN/B=1

A-187

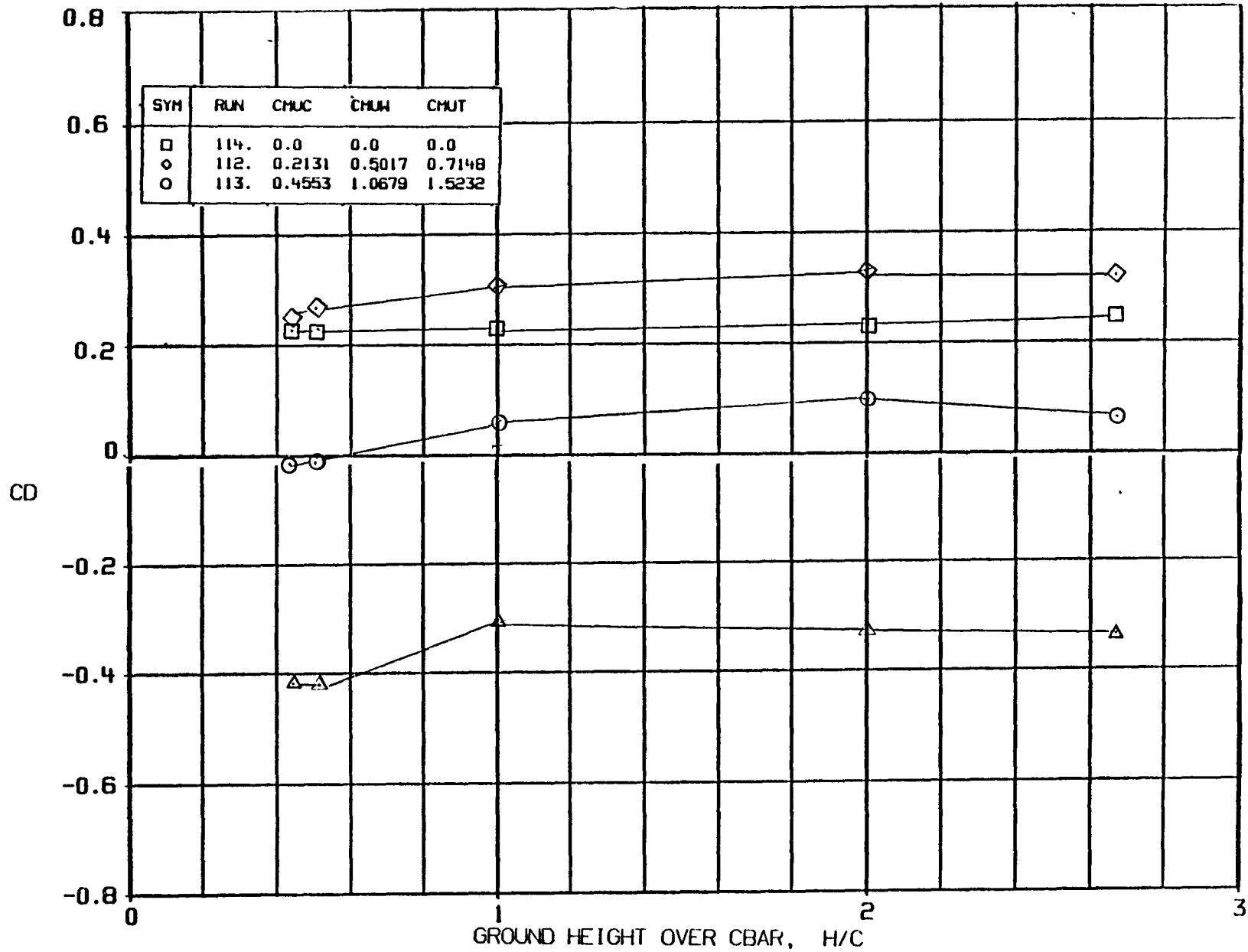


FIGURE A32b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC3W6V, DELF=45, BN/B=1

A-188

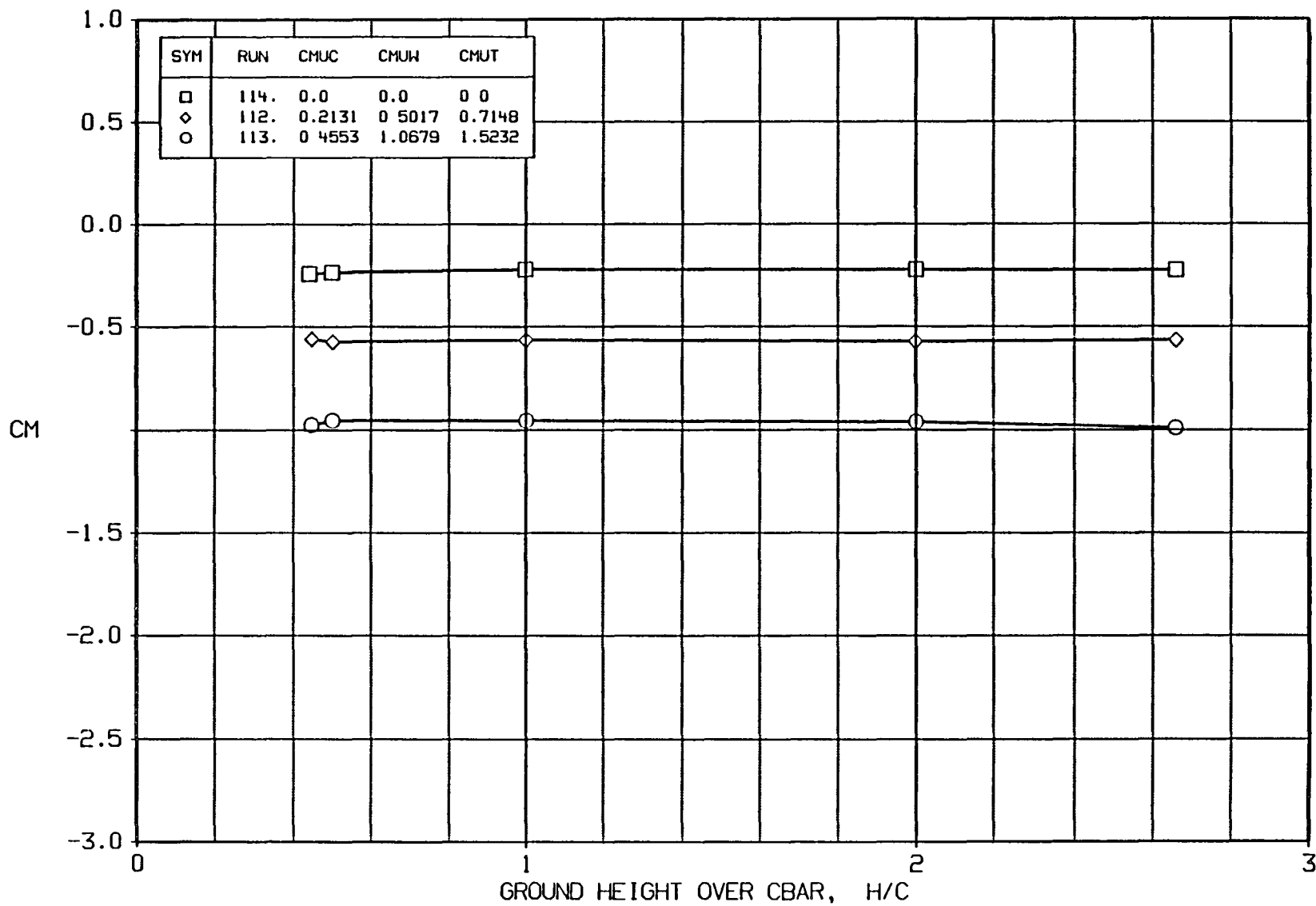


FIGURE A32c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC3W6V, DELF=45, BN/B=1

A-189

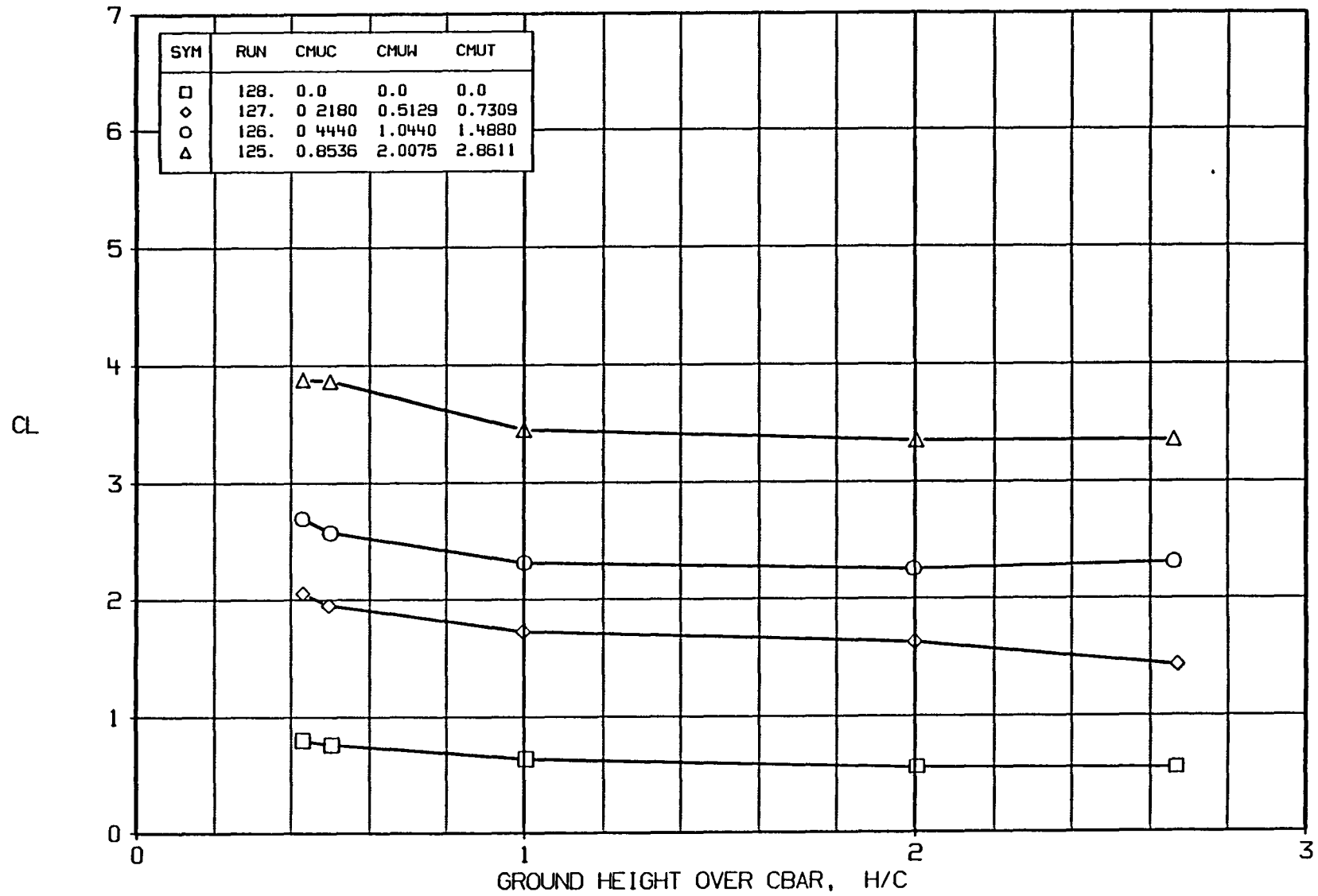


FIGURE A33a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45,  $BN/B=1$ , DELC=10°

A-190

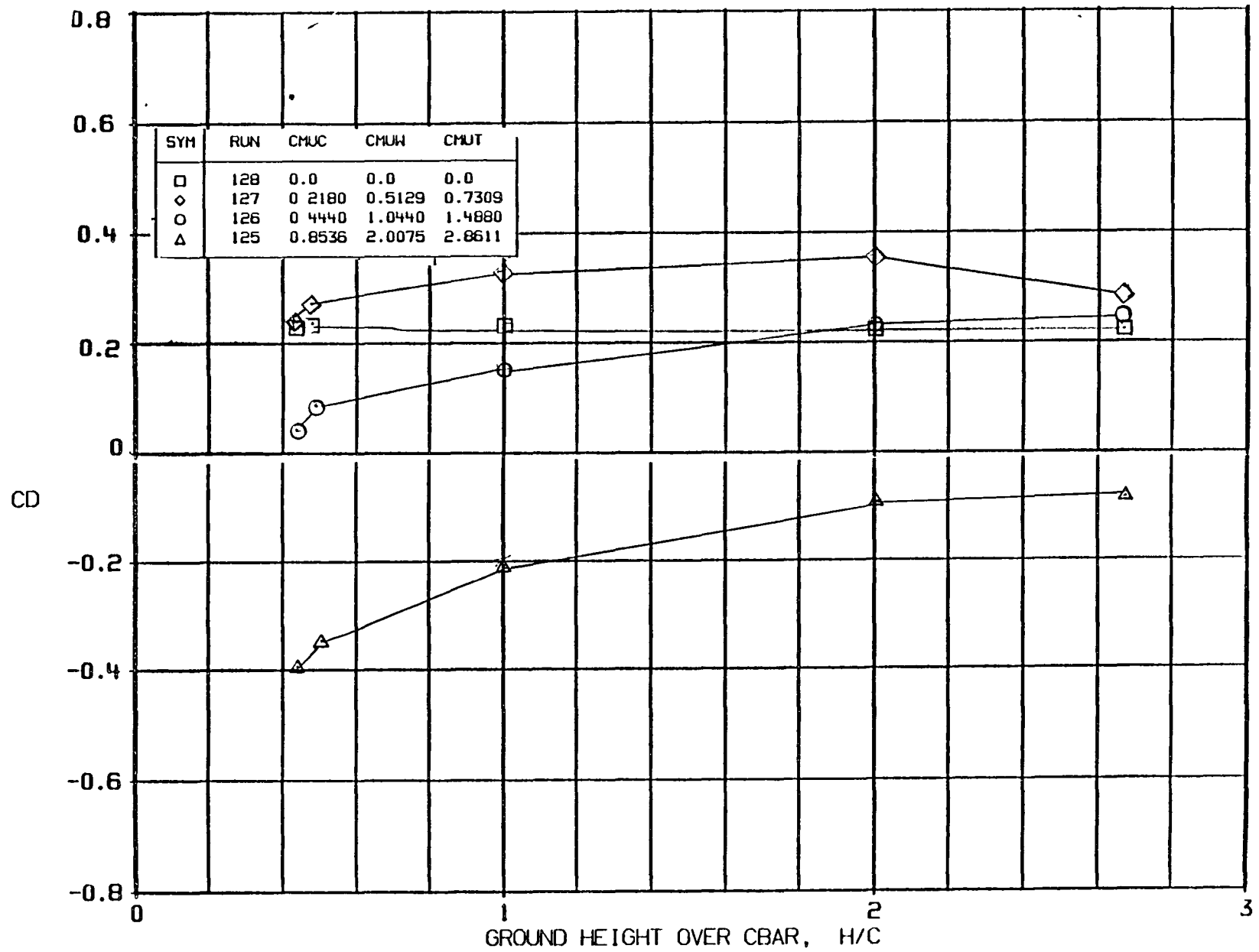


FIGURE A33b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10°



A-191

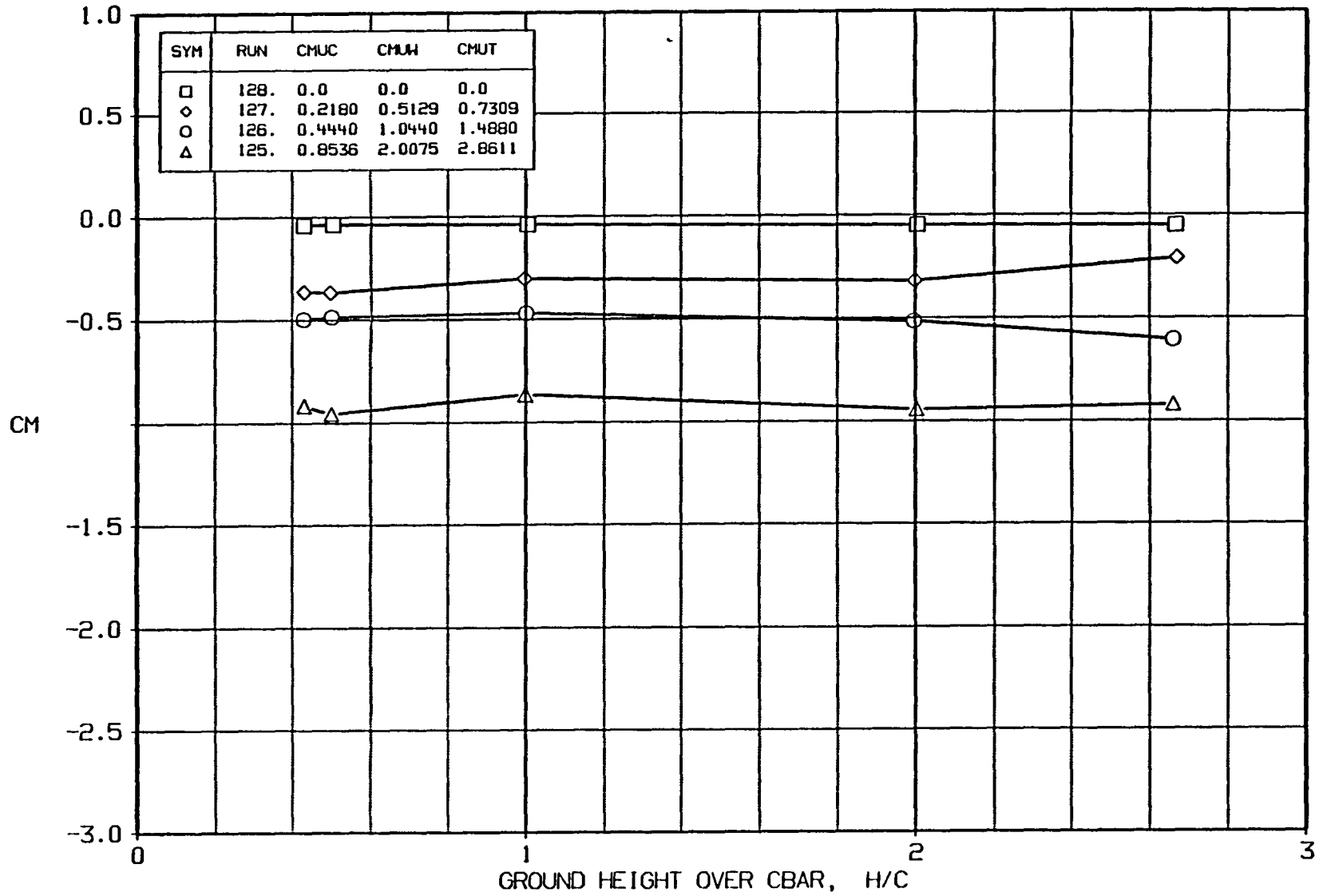


FIGURE A33c BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45,  $BN/B=1$ , DELC=10°

A-192

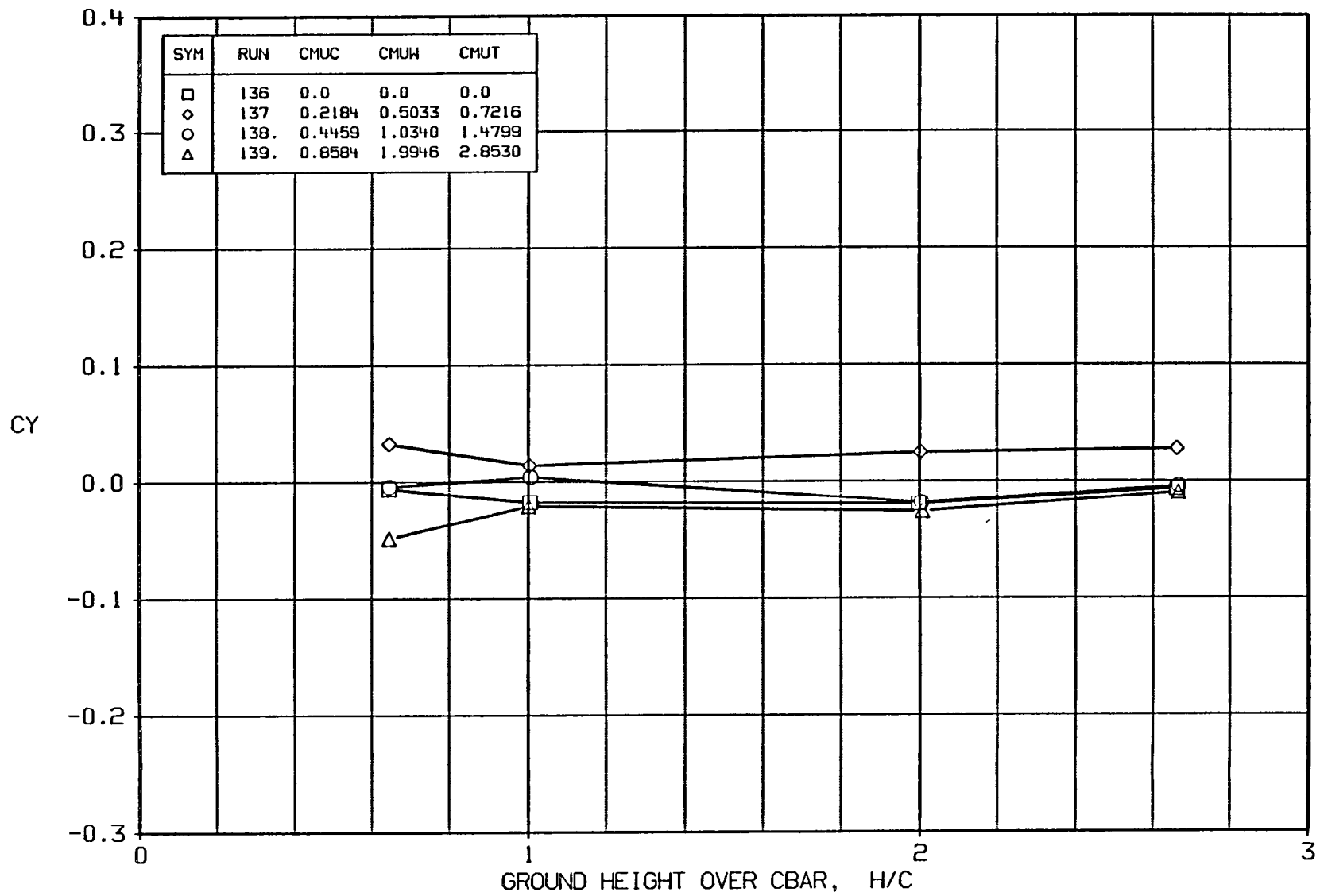


FIGURE A34a BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45, BN/B=1  $\beta = 0.8, \alpha = 2$

A-193

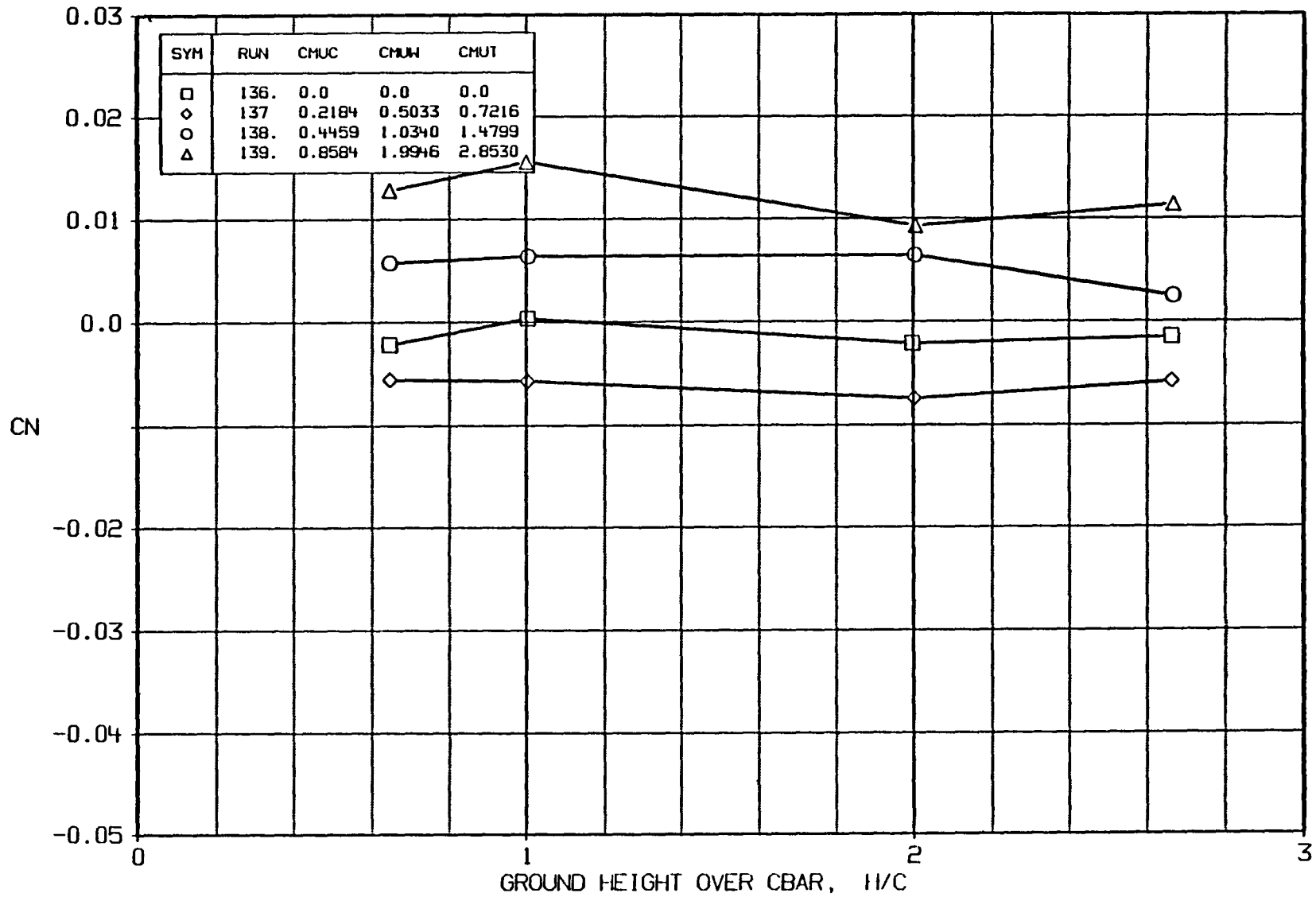


FIGURE A34b BASIC DATA EFFECT OF CMU  
CONFIGURATION BC1W6V, DELF=45,  $BN/B=1$   $\beta = 0.8$ ,  $\alpha = 2$

A-194

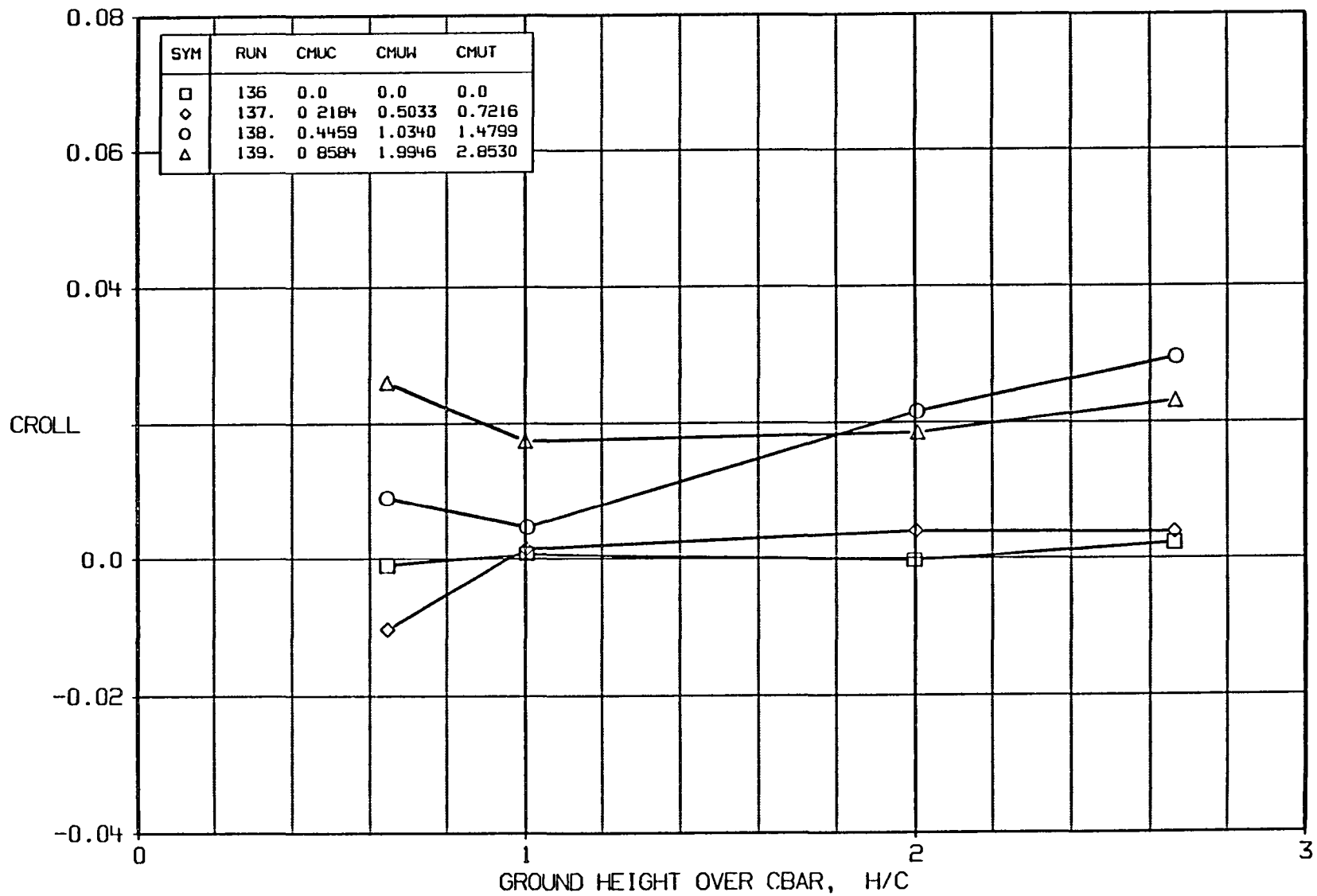


FIGURE A34c BASIC DATA EFFECT OF CMU  
 CONFIGURATION BC1W6V, DELF=45, BN/B=1,  $\beta = 0.8$ ,  $\alpha = 2$

A-195

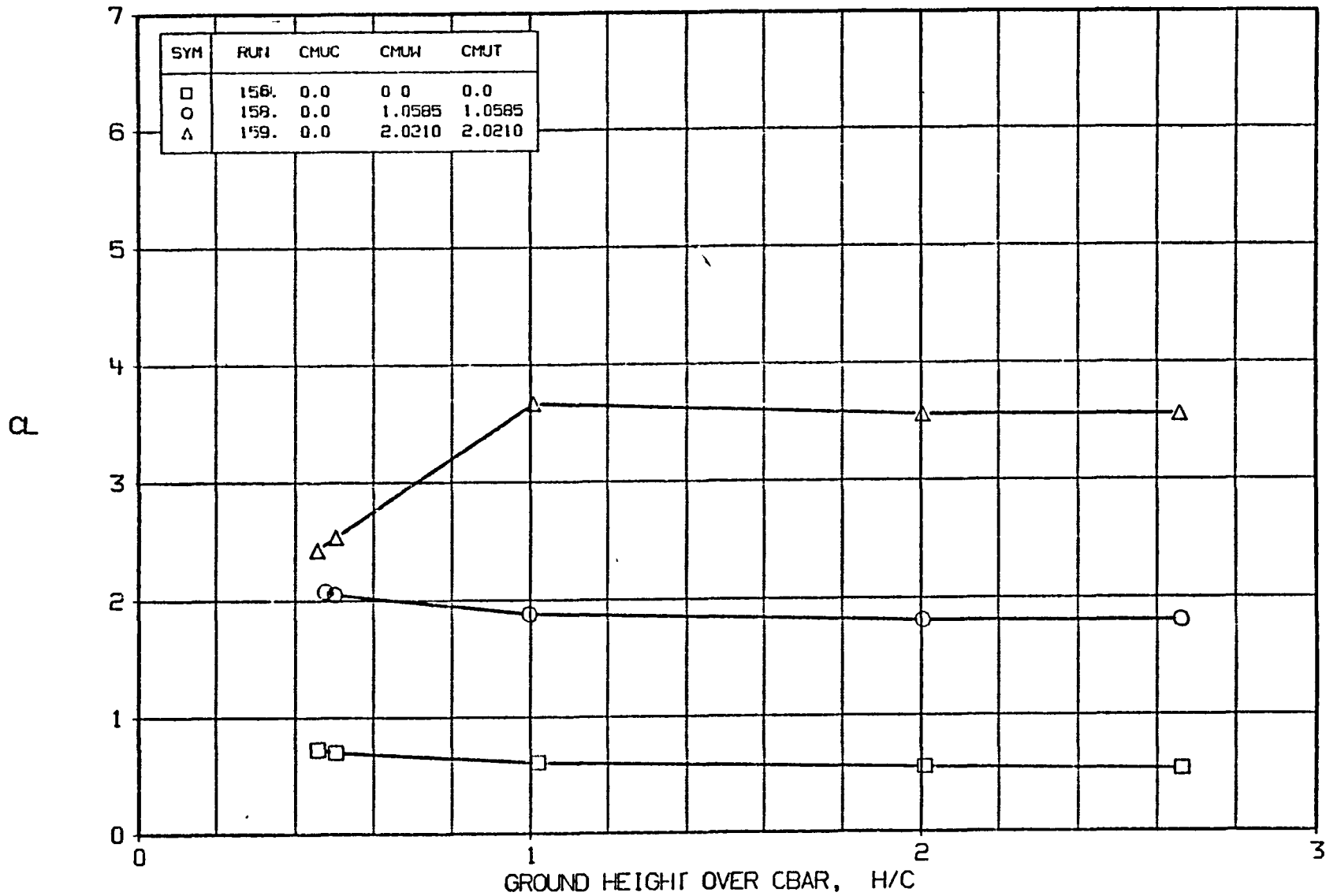


FIGURE A35a BASIC DATA EFFECT OF CMU CONFIGURATION BWSV, DELF=45, BN/B=1

A-196

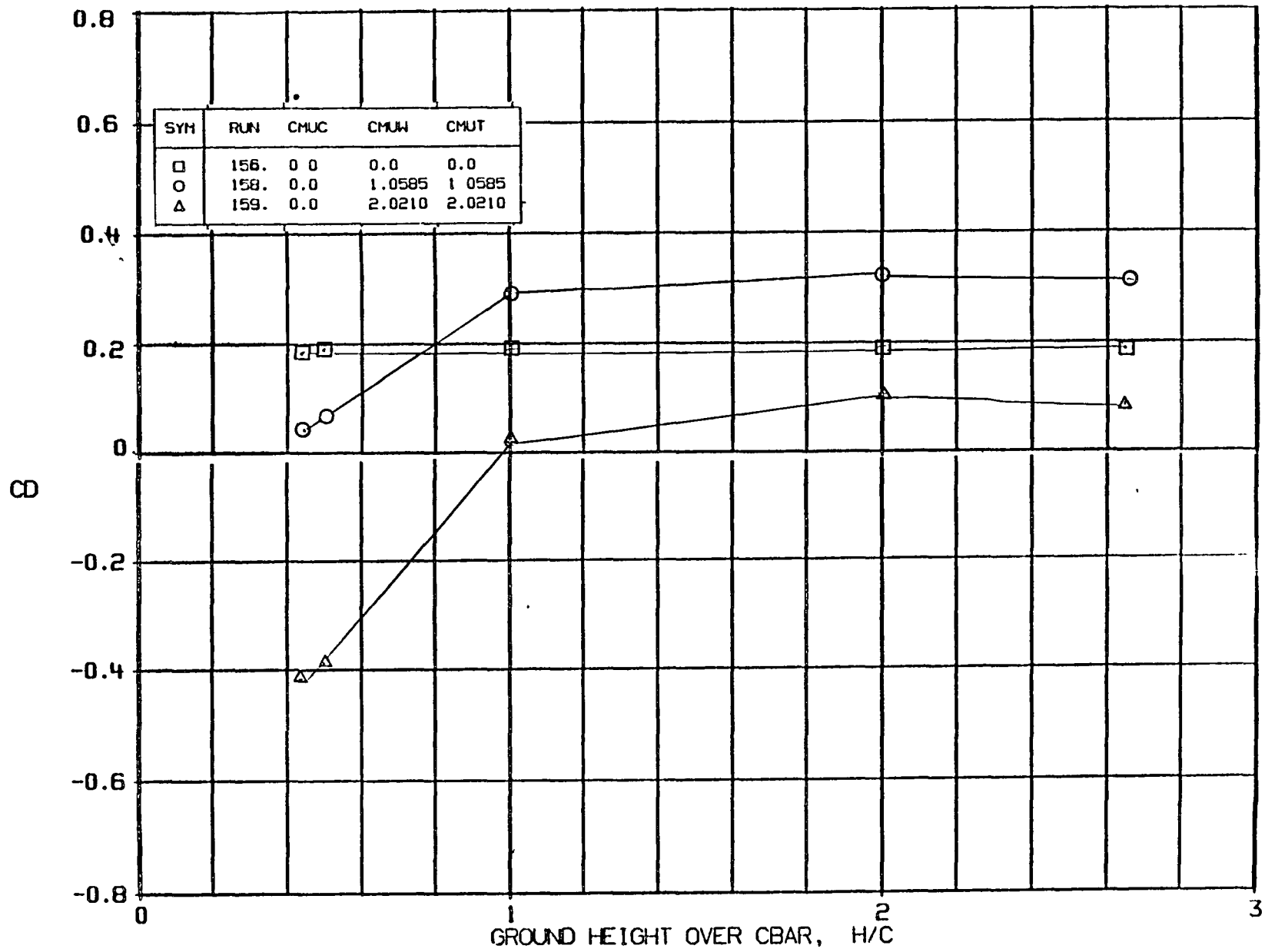


FIGURE A35b BASIC DATA EFFECT OF CMU CONFIGURATION BW6V, DELF=45, BN/B=1

A-197

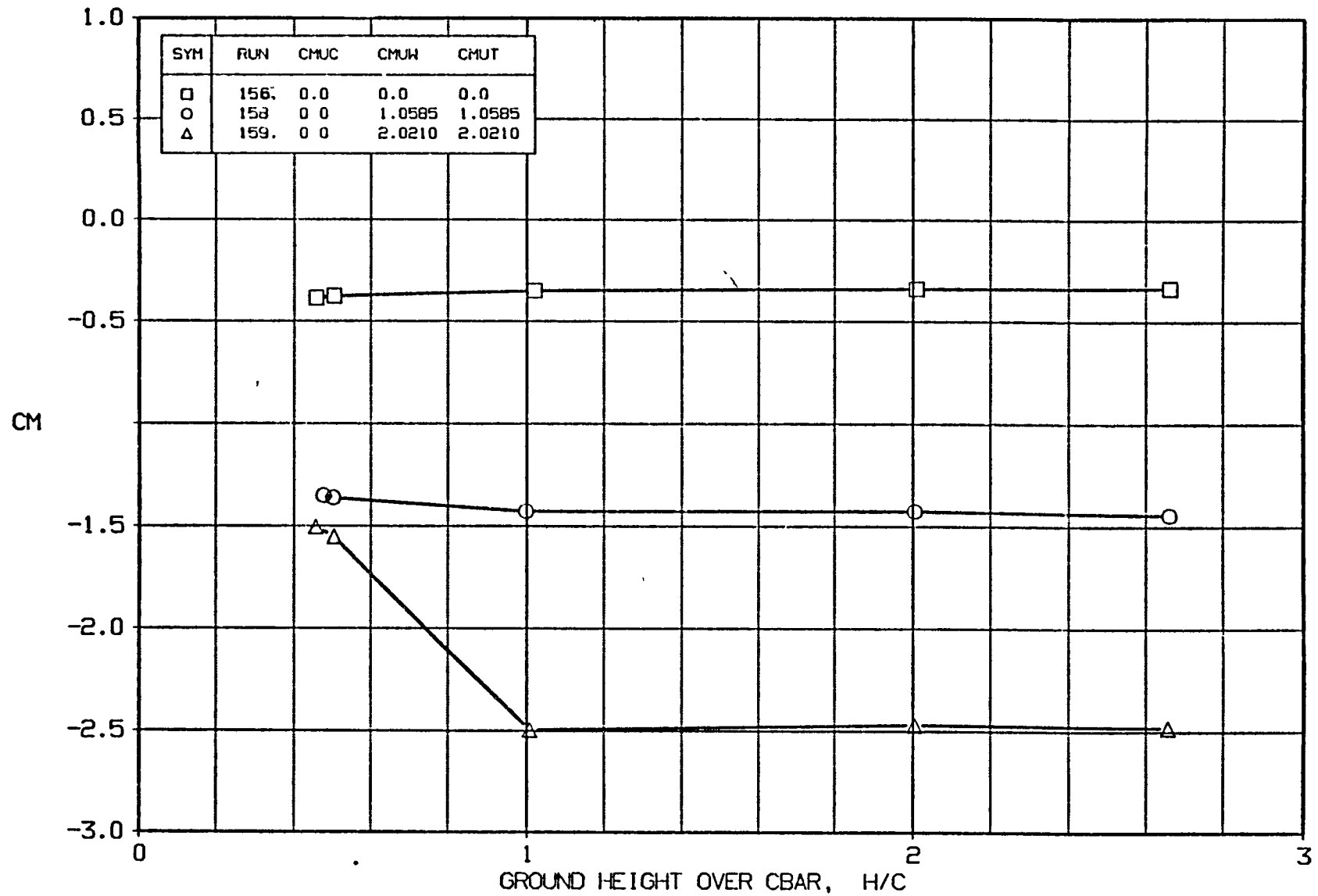


FIGURE A35c BASIC DATA EFFECT OF CMU CONFIGURATION BW6V, DELF=45, BN/B=1

A-198

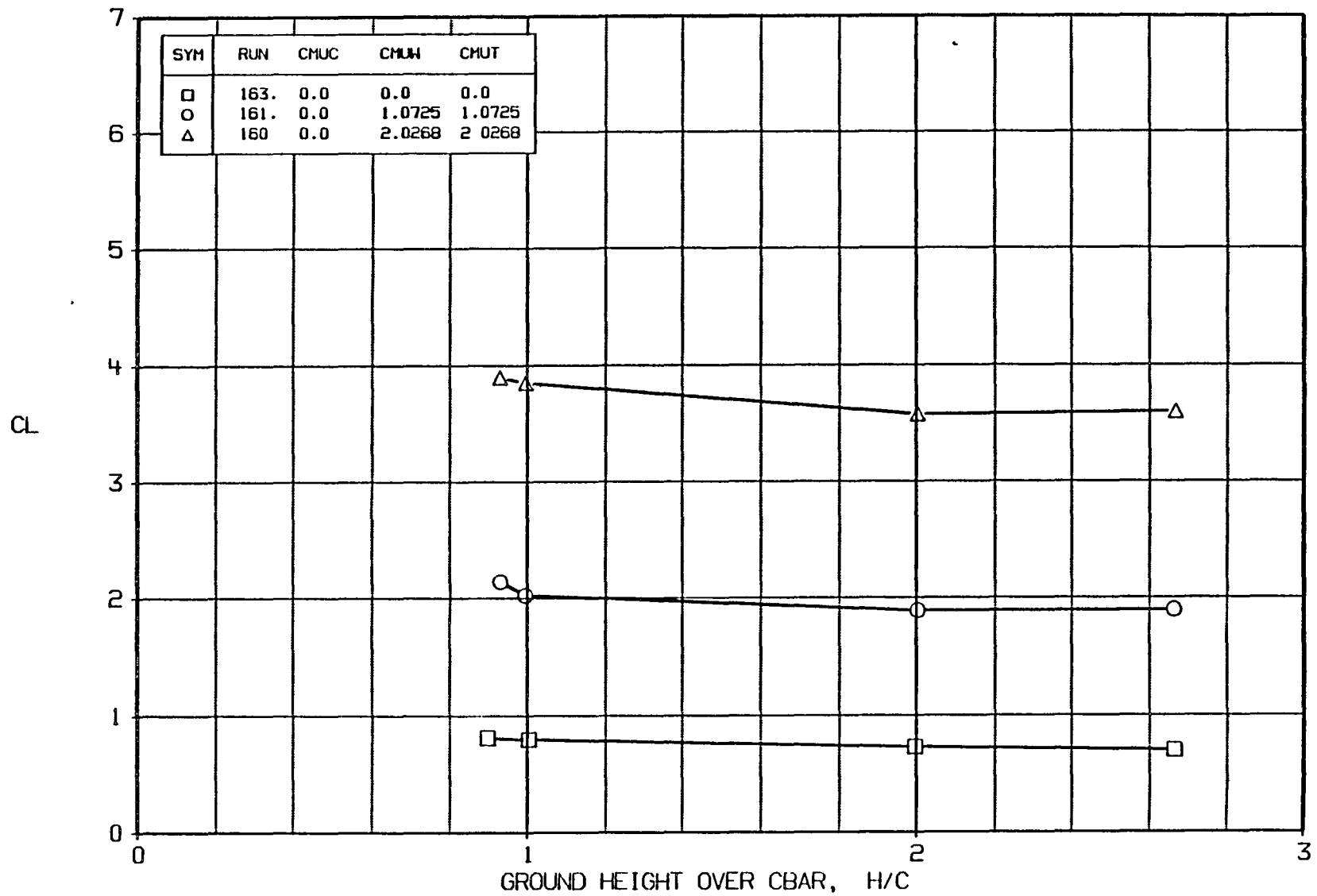


FIGURE A36a BASIC DATA EFFECT OF CMU CONFIGURATION BWSV, DELF=45,  $BN/B=1$   $\alpha = 4^\circ$



A-199

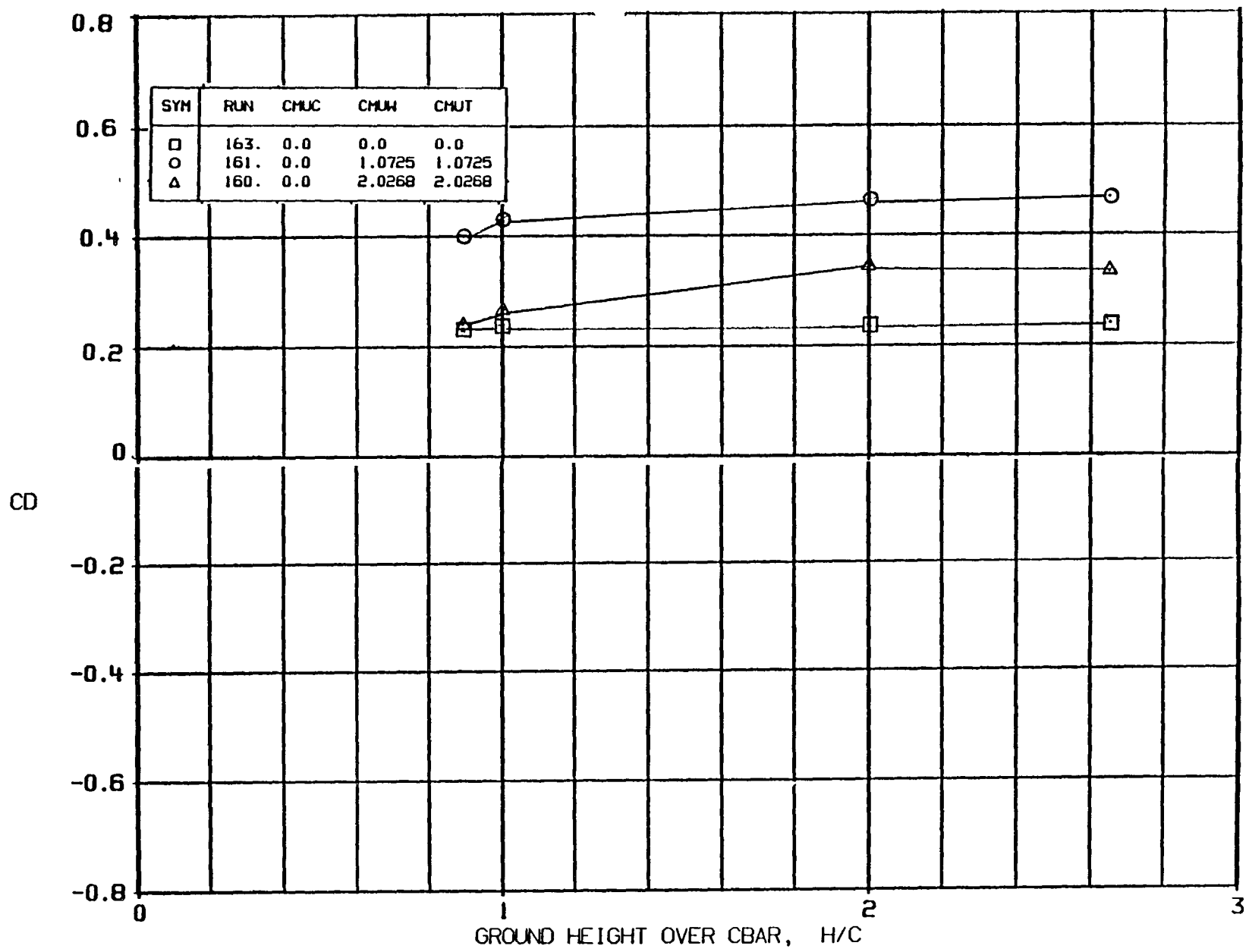


FIGURE A36b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45,  $BN/B=1$   $\alpha = 4^\circ$

A-200

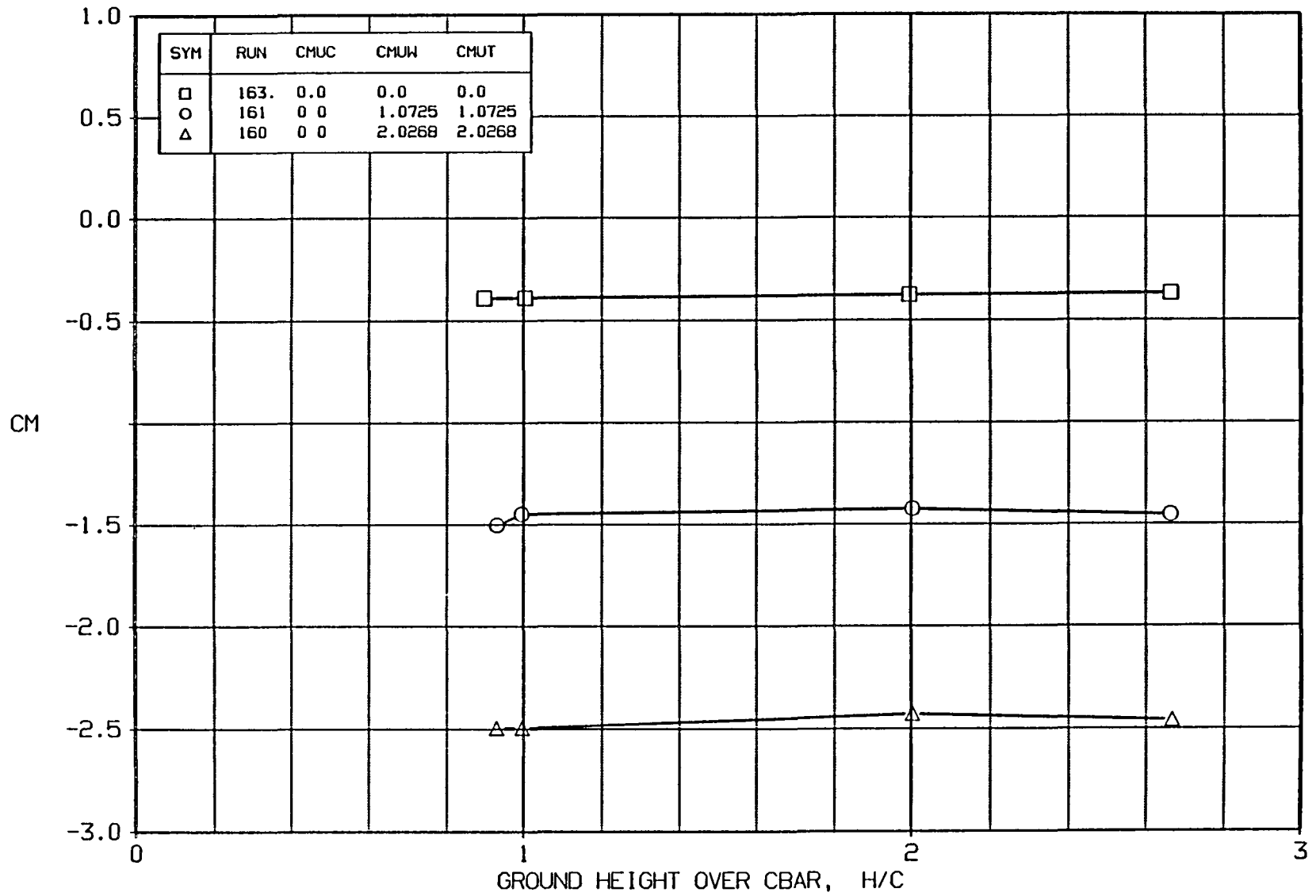


FIGURE A36c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45,  $BN/B=1$   $\alpha = 4^\circ$

A-201

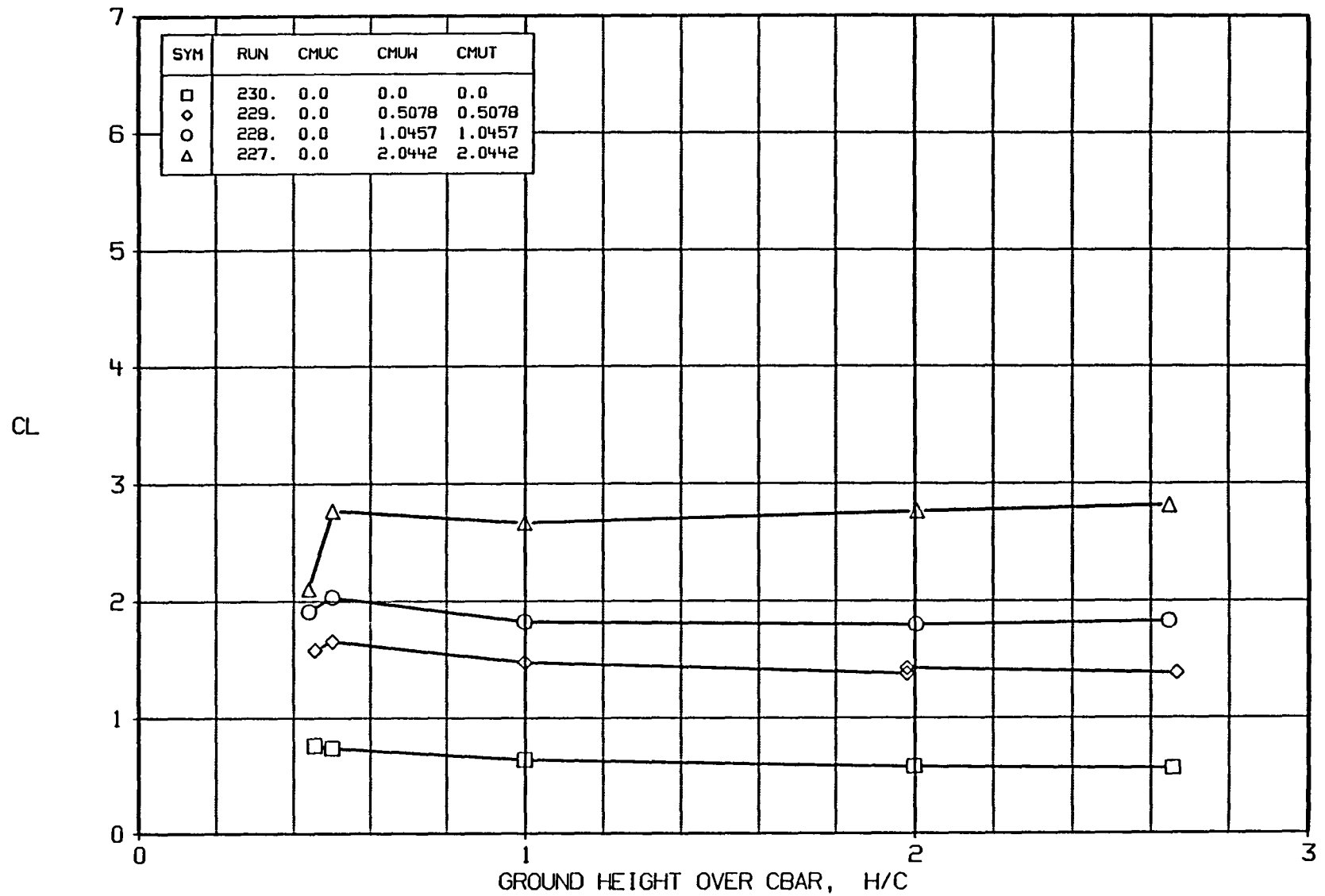


FIGURE A37a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V,  $BN/B=0.5$ , DELF=45

A-202

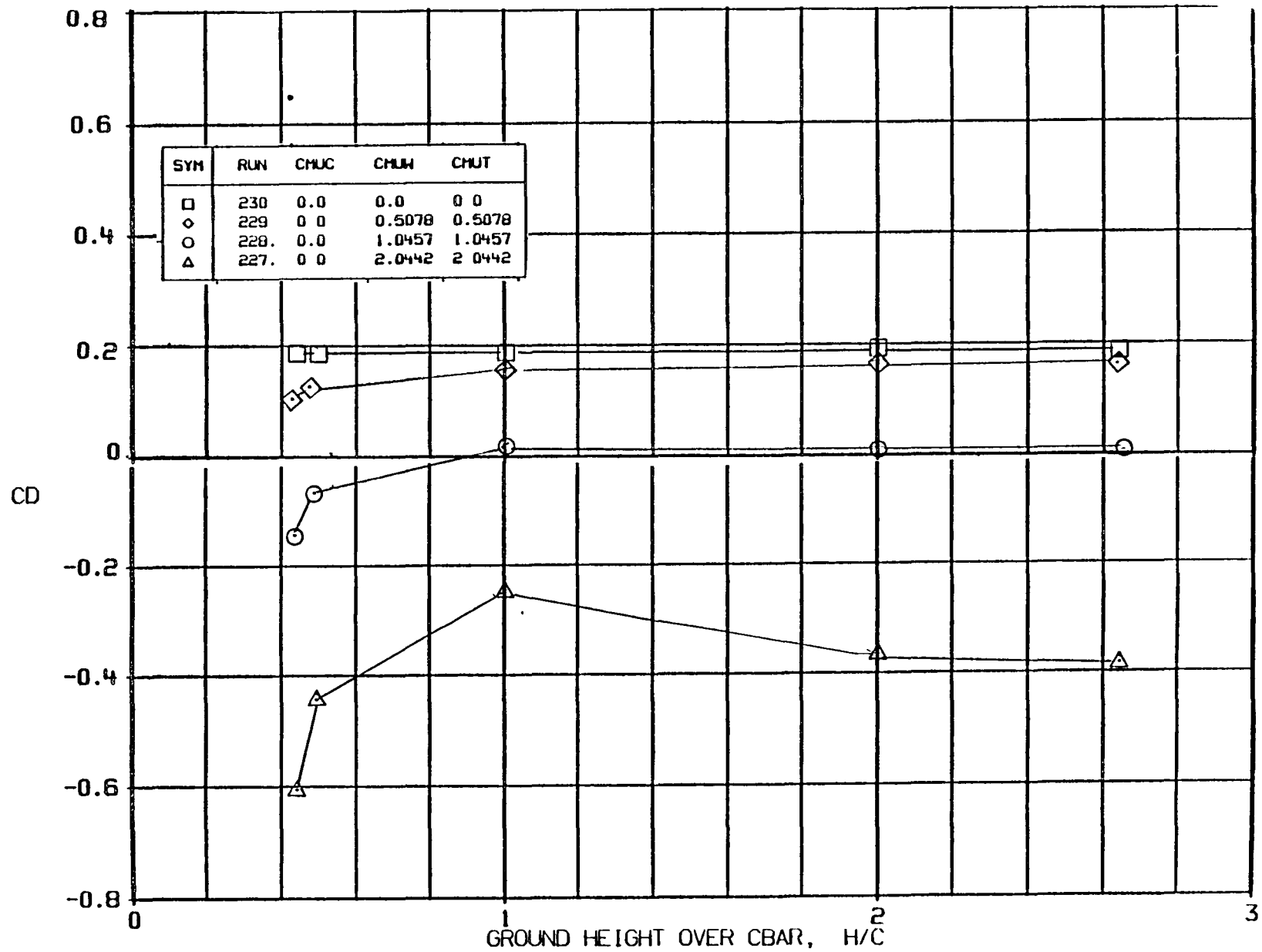


FIGURE A37b BASIC DATA EFFECT OF CMU CONFIGURATION BW6V,  $BN/B=0.5$ , DELF=45

A-203

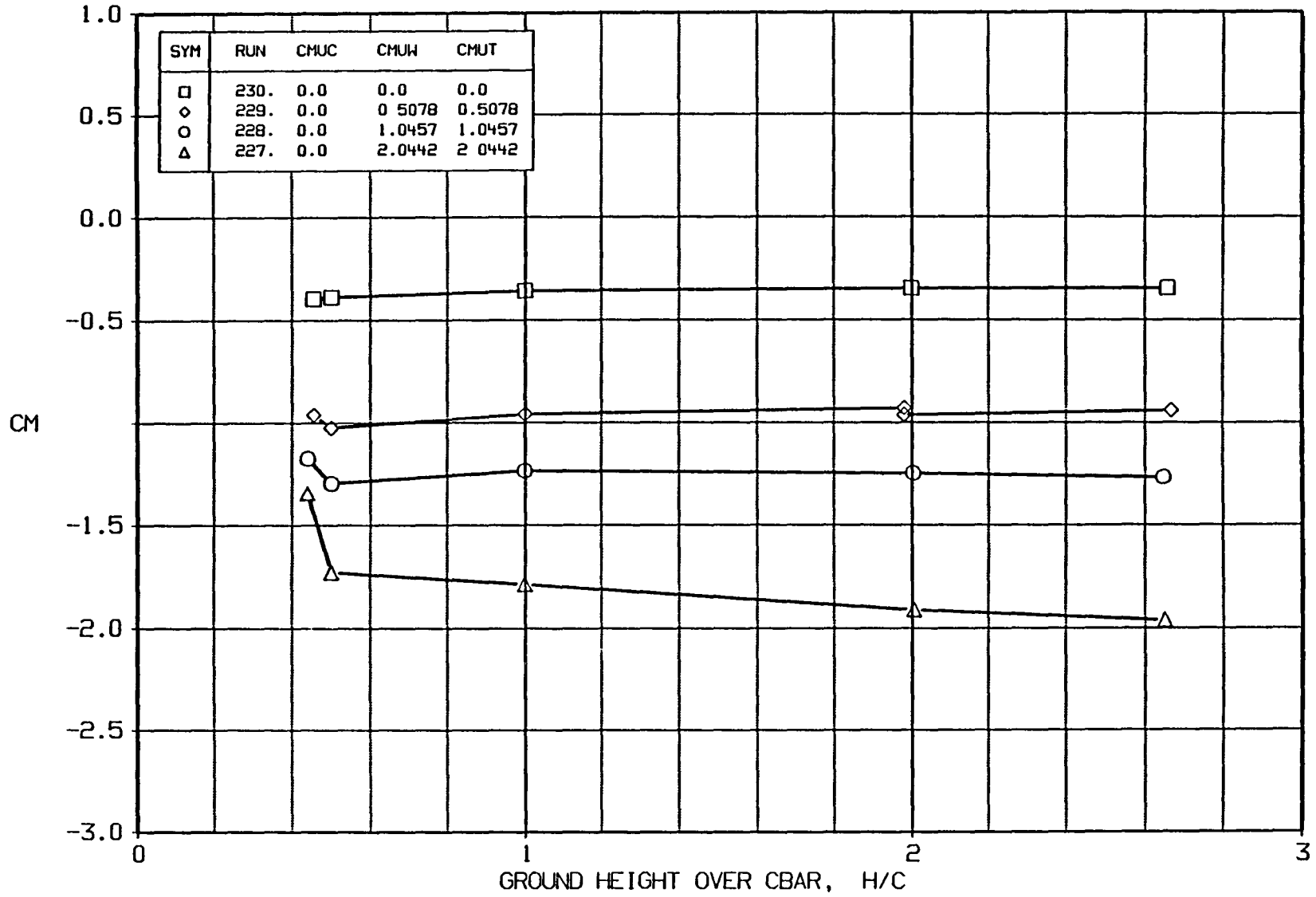


FIGURE A37c BASIC DATA EFFECT OF CMU CONFIGURATION B<sub>1</sub>6V, **BN/B=0.5**, DELF=45

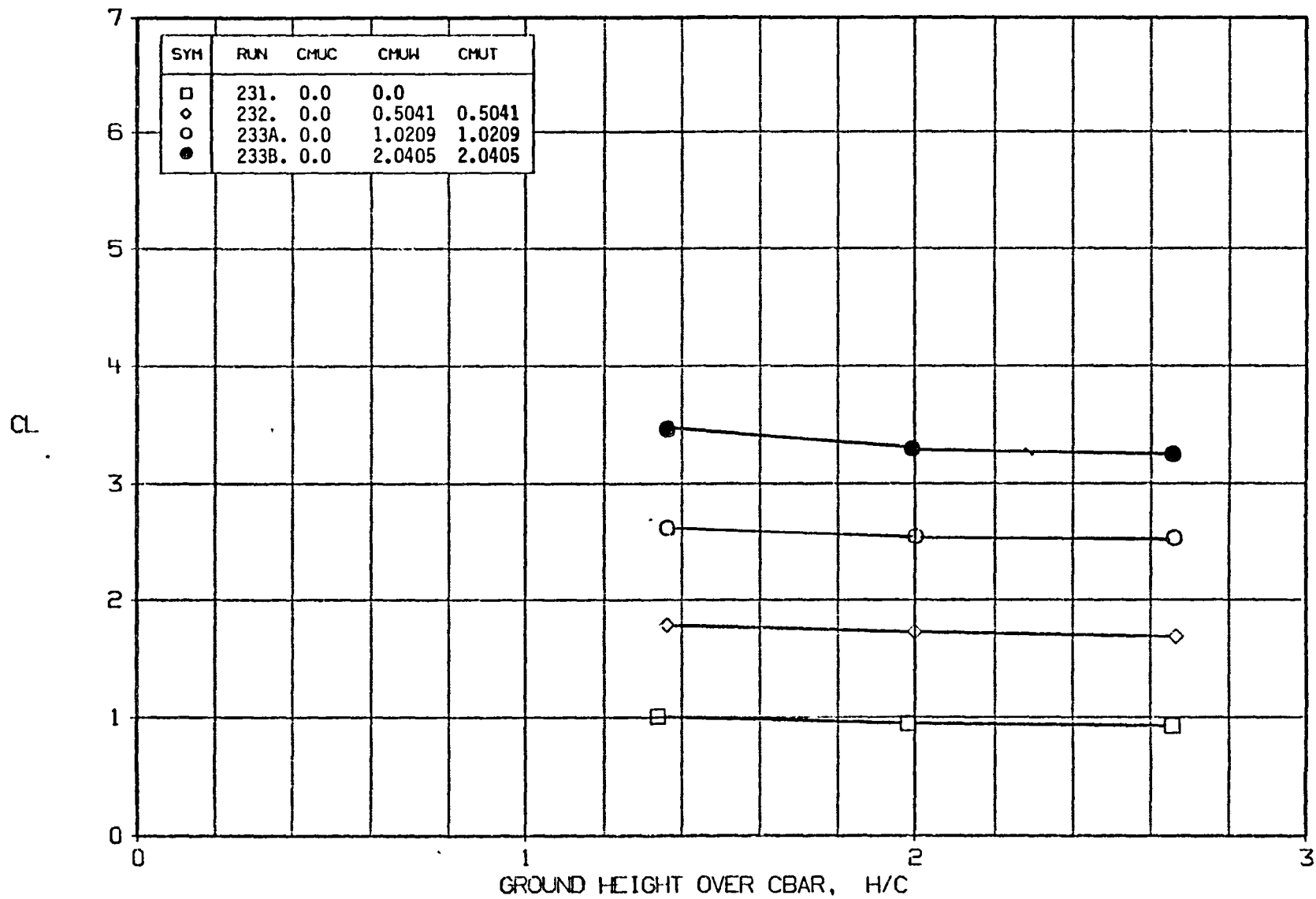


FIGURE A38a BASIC DATA EFFECT OF CMU  
 CONFIGURATION BW6V,  $BN/B=0.5$

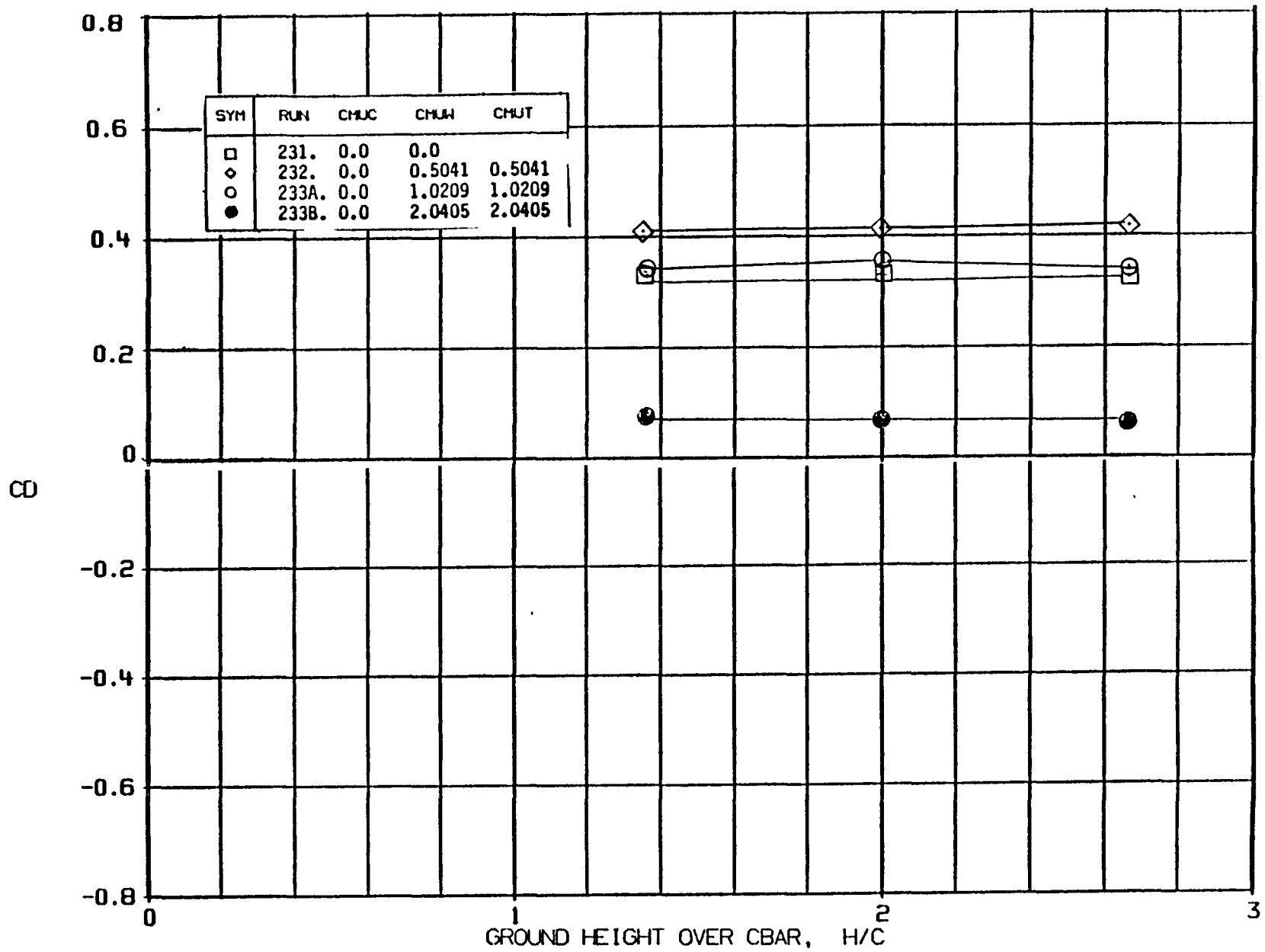


FIGURE A38b BASIC DATA EFFECT OF CMU  
CONFIGURATION BWSV,  $BN/B=0.5$

A-206

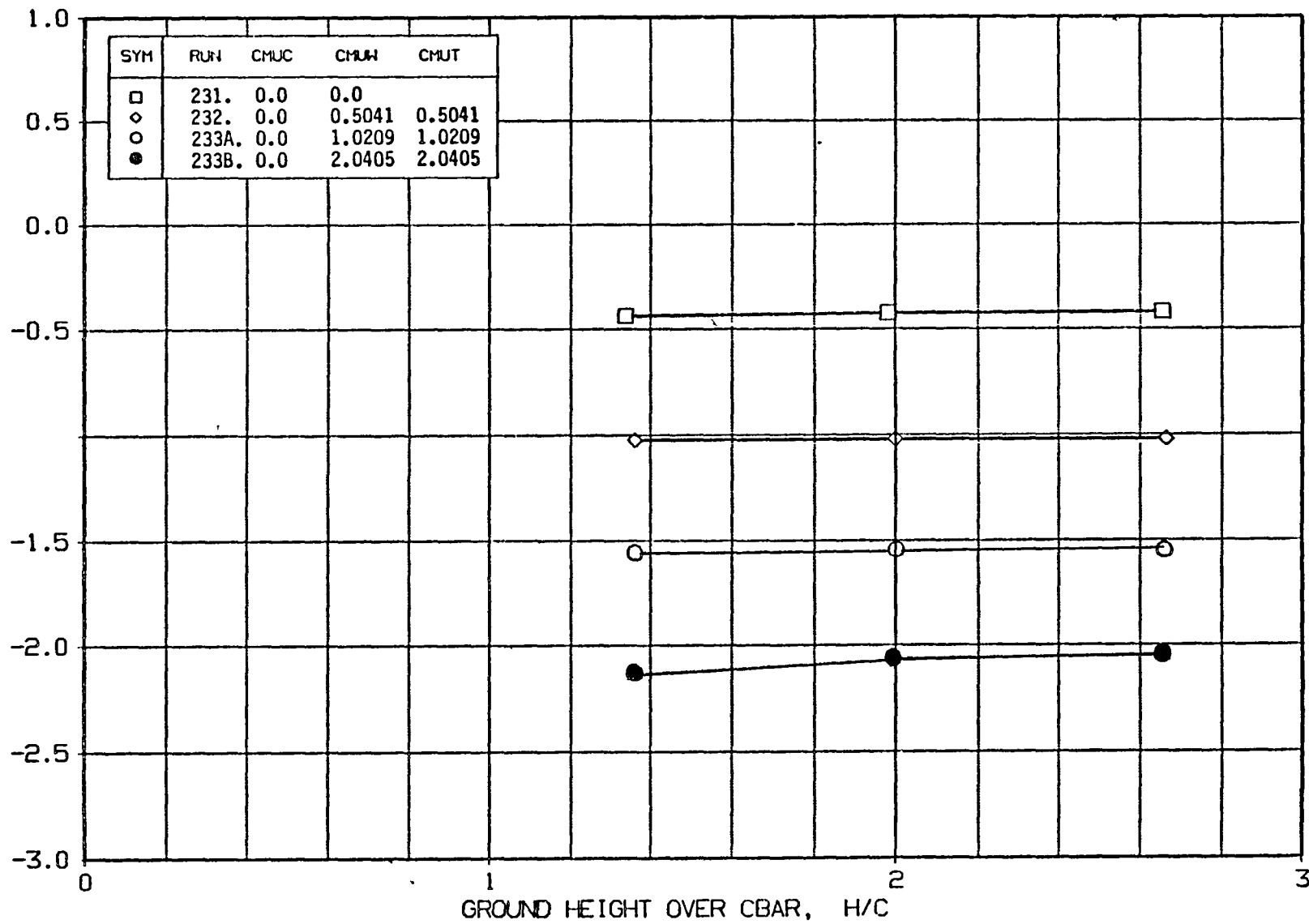


FIGURE A38c BASIC DATA EFFECT OF CMU CONFIGURATION BW6V,  $BN/B=0.5$



A-207

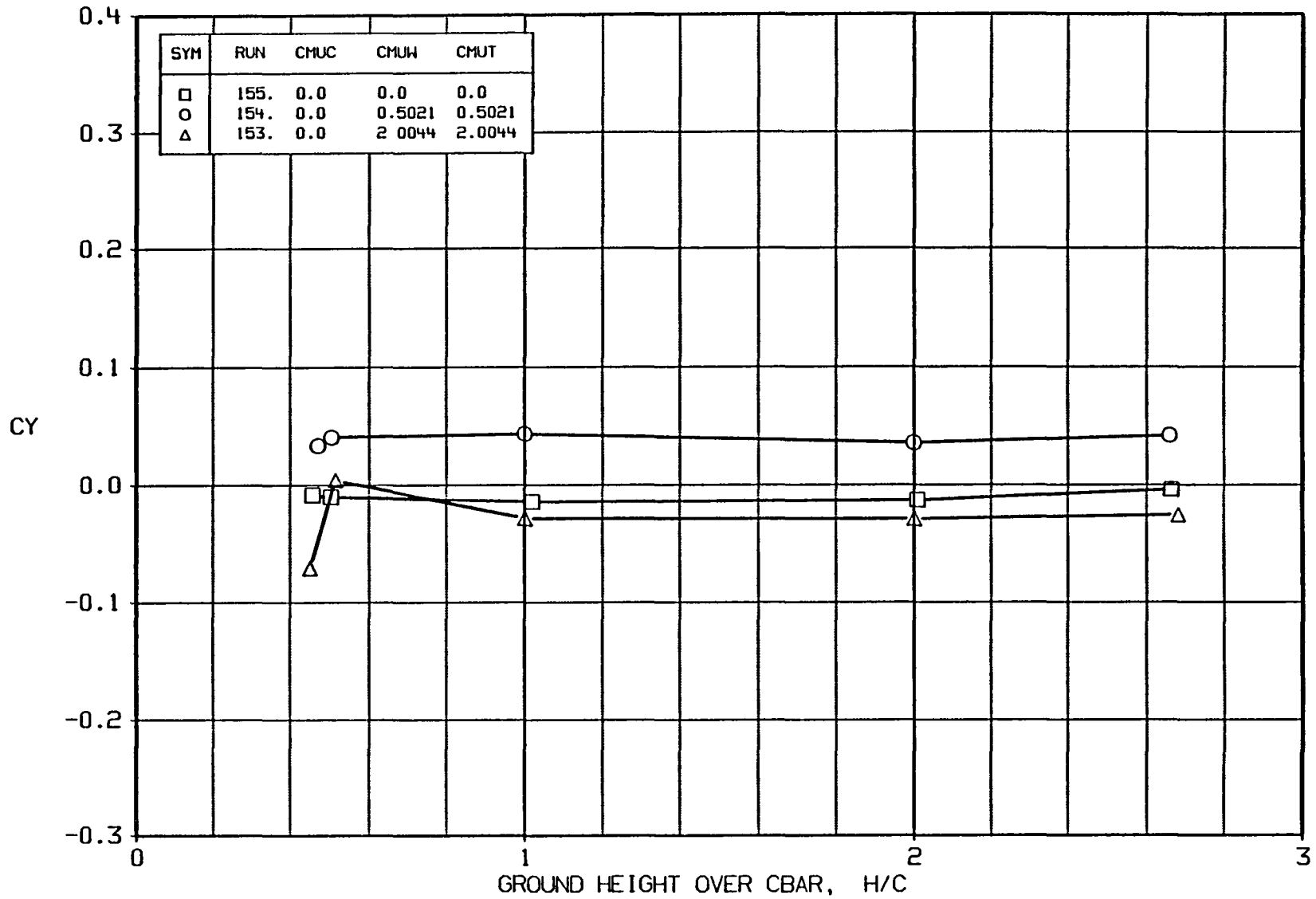


FIGURE A39a BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, **BN/B=1**,  $\beta = 0.8$

A-208

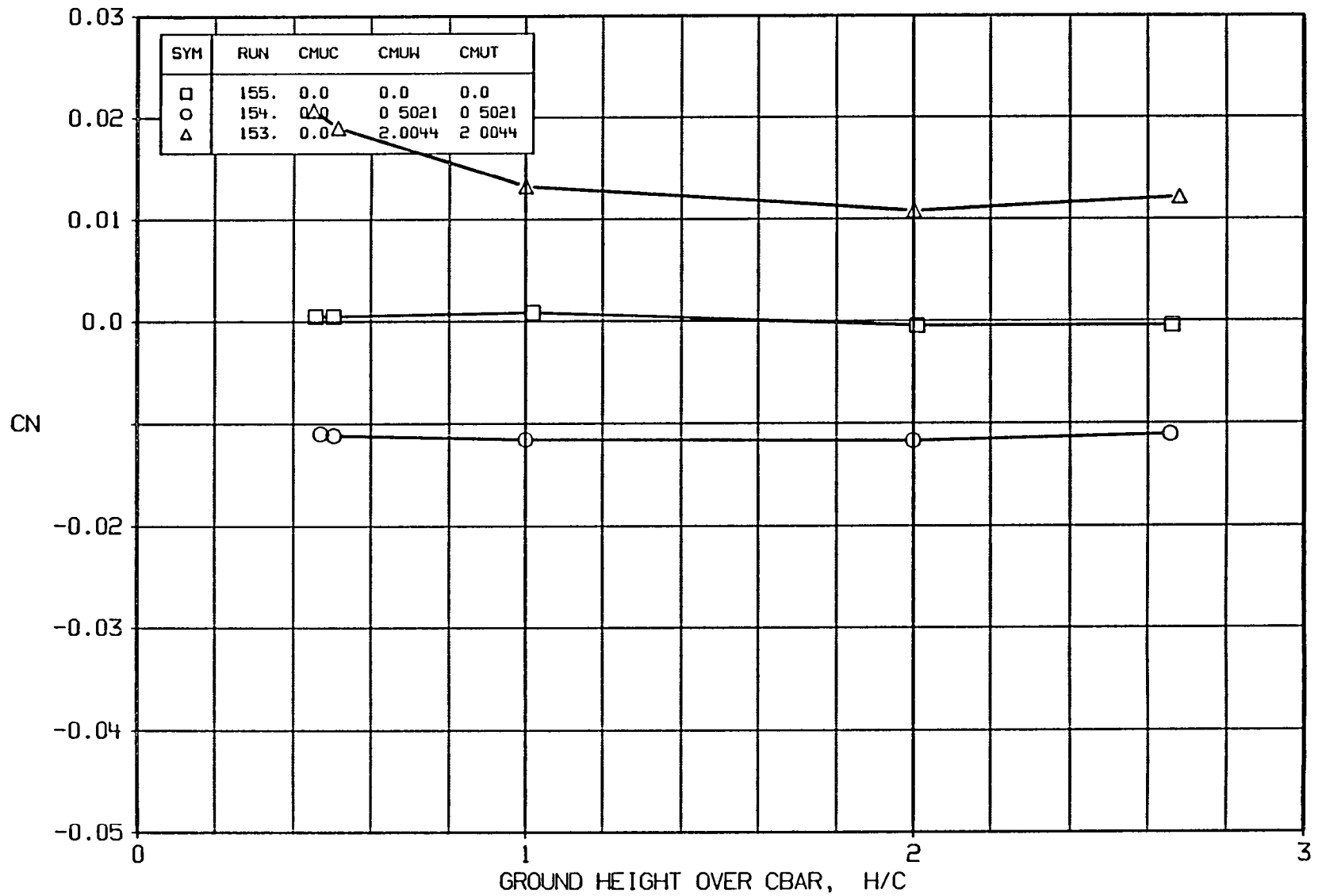


FIGURE A39b BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45, **BN/B=1**,  $\beta = 0.8$

A-209

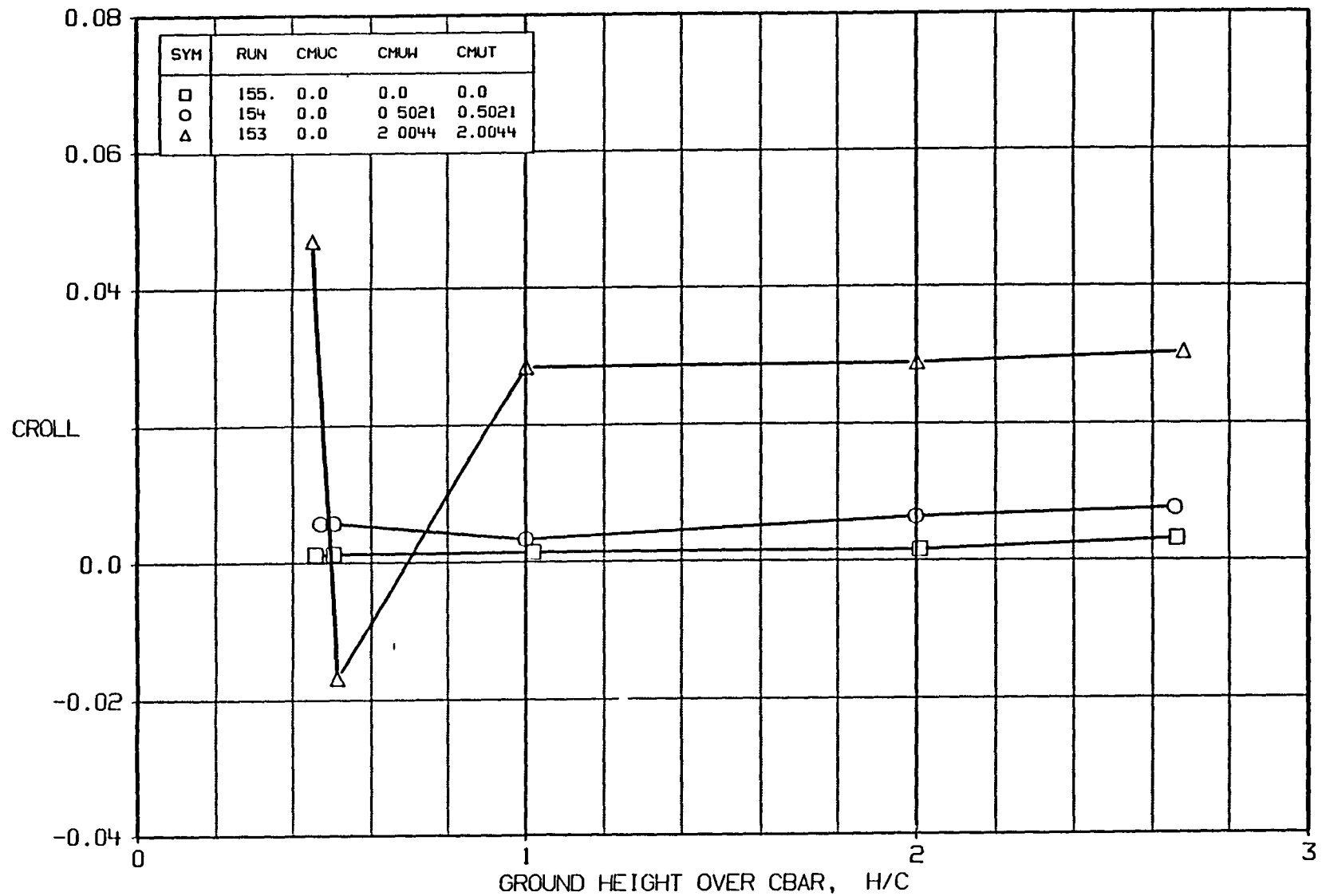


FIGURE A39c BASIC DATA EFFECT OF CMU  
CONFIGURATION BW6V, DELF=45,  $BN/B=1$ ,  $\beta = 0.8$

### A3.0 TABULATED SURFACE PRESSURE DATA

The surface pressure distribution for selected data runs is presented in Tables A1 to A50. The pressure coefficients are presented for the canard and wing with the locations described for each coefficient. The fuselage pressures are presented for various fuselage stations three inches to the left of the plane of symmetry. The wing pressures, presented as miscellaneous pressure coefficients, are as described below:

#### MISCELLANEOUS WING PRESSURES

	X/C	BP	
1	0.33	9	Upper
2	0.33	9	Lower
3	0.33	19	Upper
4	0.33	19	Lower
5	0.33	23.5	Upper
6	0.33	23.5	Lower
7			
8			
9	Downwash probe pressure		
10	coefficients		
11			
12			

TABLE A1. RUN61, BW6V, DELF=0, CMU=1.0, (BN/B)=1

RUN SEQ		PROPLULSIVE								WING / CANARD PRESSURES			PAGE 74
61 4	CLAERD =	0 3119	CDAERD =	0.0606	DCLC =	0.0	DWLC =	0.2681	BASEPR =	8.1904			
	ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
	0.06	2111.83	2099 73	12 09	101 17	0.01	2 67	0 0	1.032	1 032	87.2626	0.644190E+06	
	BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
	-0.01	0 58	-0 94	-0 34	0 01	-0 01	0.02	0.30	-0.27	0.30	0.03		

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO	CP
0 0	-0 2375	-0 2375	0.0	0.5167	0 5167	0 2890	0.3744	0.3744	38.2	-0.1259	1	-0.3161
2 5	-0 2172	-0 2375	2 5	-0 1949	-0 2802	-0 2802	-0.5080	-0.3657	43.4	-0.1043	2	-0.1515
5 0	-0 2172	-0 2375	5 0	-0 1664	-0 2233	-0 3087	-0.5080	-0.4226	50 4	-0.1043	3	-0 3983
10 0	-0 2172	-0.2375	10 0	-0.2518	-0.2802	-0 3657	-0.3941	-0.4226	56 1	-0.1043	4	-0 1515
15 0	-0.2375	-0 2375	15 0	-0 2802	-0 3372	-0.3657	-0.3941	-0 4510	62.5	-0.2343	5	-0 3161
25 0	-0 2172	-0 2375	24.0	-0.2802	-0.3372	-0 3657	-0 3941	-0 3941	69.7	-0.2126	6	-0.1515
35 0	-0 2172	-0.2172	33 0	-0.3161	-0 3161	-0.3983	-0.4807	-0.3161	76.9	-0.1693	7	0 5069
50 0	-0.2172		54 0	-0 3161	-0.3983	-0 4807	1.0006	-0 3983	83.4	-0.1476	8	-0 2161
56 0	-0 2172	-0 2375	65.0	-0 4942	-0 5375	-0 5592	-0.5808	-0.5592	89.4	-0.2126	9	-0 0221
65.0	-0.2172	-0.2375	78 5	-1.1007	-1.1440	-1 1657	-1.2307	-1.3174	95.4	-0.2559	10	0 5950
76 0	-0 2172	-0.2578	79.5	130 2903	130.2903	130 2903	130.2903	124.2206	101.1	-0.2126	11	0.2953
79 0	-0.2375	-0.2375	80.5	-54.0155	-45.9406	-58 8084	-53.8989	-40.7963			12	0.0131
80.5	-0.2158	-0.2012	81.3	-21.4959	119.6643	-5.6529	-10.5185	-3.0883	38.2	-0.1043		
81 0	-0.2012	-0 1865	82.0	28.2891	25.7538	20 0529	2.9351	2 1437	43.4	-0.0609		
82 0	-0 2158	-0.2012	84 0	14.3513	8.2990	1 0445	1.5428	1.1032	50.4	-0.1043		
84 0	-0.2012		87.0	3.4920	-2.0624	-1.4909	1.1032	0.4290	56.1	-0.0609		
87.0	-0.2012		89.0	0.1190	-0.0750	-1 9616	-1.5914	1.1593	62.5	-0.0609		
89.0	-0 2343	-0.2343	93 0	0.3305	0.4716	-0.0045	-0.3395	-0.5864	69.7	-0.0609		
93.0	-0 2343		96 0	0 2600	-0.5687	1.2298	1.5824	-1.5737	76 9	-0.0826		
96 0	-0 2343	-0 2343	100 0	-2.3319	-2.2438	-1.7853	-1 6619	-1 6796	83.4	-0 1043		
100 0	-0.2343	-0 2126	96.0	-0.0397	-0.1455	-0 1102	-0.1455	0 0131	89.4	-0.0609		
96.0	-0.2343	-0.2126	84 0	0 0837	0 1013	0 2777	0.1013	0 0661	95.4	-0.0393		
84 0	-0.2343	-0 2126	73 5	-0 2338	-0.0692	-0.0692	-0.0692	-0.0692	101.1	-0 0609		
73.0	-0.2343	-0 2126	54.0	-0.1515	-0 1515	-0.1515	-0 1515	-0 1515				
50.0	-0.2343	-0 2126	33 0	-0 1515	-0.1515	-0 1515	-0 1515	-0 1515				
35.0	-0.2172	-0.2172	24.0	-0 1379	-0 1379	-0 1664	-0.1095	-0.1949				
25.0	-0 2172	-0.2375	10.0	-0.0810	-0.0810	-0.0810	-0.1095	-0.0810				
10 0	-0.2375	-0.2172	5.0	-0 0810	-0.1095	-0.0810	0.0044	-0 1515				
5.0	-0 2375	-0.2375	2.5	-0 0241	0.0044	-0.0526	0 0897	-0.0692				
2 5	-0.2375	-0.2375										

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TABLE A1. (Continued)

RUN SEQ	PROPLULSIVE										WING / CANARD			PRESSURES			PAGE 76	
61 6	CLAERO =	0 5431	CDAERO =	0 1122	DCLC =	0.0	DWLC =	0.3440	BASEPR =	8 1904								
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN							
4 06	2112 04	2099 94	12.09	101 14	0 01	2 66	0 0	1.053	1 053	86.8984	0.644619E+06							
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	COTR								
-0 01	0 89	-0 88	-0 40	0 01	-0 01	0 01	0 54	-0.33	0.53	0 07								
***** CANARD *****			***** WING *****						**FUSELAGE**			**MISC.**						
%X/C	BP=3 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----		XLOC	CP	NO.	CP				
0 0	CP	CP	0 0	CP	CP	CP	CP	CP	38.2	-0.1043	1	-0.4807	1	-0.4807				
2 5	-0 2172	-0 2172	2 5	-0 0810	-0 7072	-1.1911	-1 1627	-1 3619	43.4	-0.0826	2	-0.0692	2	-0.0692				
5 0	-0 2172	-0 2172	5 0	-0 5649	-0 7072	-0 9065	-1.0488	-1.0488	50.4	-0.1043	3	-0.5630	3	-0.5630				
10 0	-0 2172	-0 2172	10 0	-0 5080	-0 6787	-0 7642	-0 8210	-0 8210	56.1	-0.1043	4	-0.0692	4	-0.0692				
15 0	-0 2172	-0 2172	15 0	-0 4510	-0 5364	-0 6503	-0 6503	-0 7356	62.5	-0.2125	5	-0.4807	5	-0.4807				
25 0	-0 2172	-0 2172	24 0	-0 4226	-0 4510	-0.5364	-0.5364	-0.6218	69.7	-0.2125	6	-0.1515	6	-0.1515				
35 0	-0.2172	-0.2172	33 0	-0 4807	-0 4807	-0.4807	-0.5630	-0.5630	76 9	-0 2125	7	0 5069	7	0 5069				
50 0	-0 1969		54 0	-0 4807	-0 4807	-0 5630	1 0006	-0 5630	83 4	-0.1909	8	-0 2161	8	-0 2161				
56 0	-0 2172	-0 1969	65 0	-0 5592	-0.6242	-0 6025	-0 6458	-0.6892	89.4	-0.2342	9	-0.0750	9	-0.0750				
65 0	-0 1969	-0 2172	78 5	-1.1224	-1 2091	-1 2524	-1.2957	-1 4474	95.4	-0 2558	10	0 5421	10	0 5421				
76 0	-0 1969	-0 2172	79 5	130 3233	130 3233	130.3233	130.3233	125.6016	101 1	-0 2125	11	0 3129	11	0 3129				
79 0	-0 2172	-0.1969	80 5	-54 7631	-47 0545	-59 2921	-54 4703	-42.1450	---BOTTOM---		12	-0 0045	12	-0 0045				
80 5	-0 2305	-0 2158	81 3	-22 1550	98 0485	-4 6858	-7.7193	-3 2935	38 2	-0.0826								
81 0	-0 2158	-0 2158	82 0	28 5818	27 3216	20 1555	-0.6408	1.5428	43.4	-0.0393								
82 0	-0 2158	-0.2305	84 0	12 5783	12 0797	2 1144	-2 5167	1 4403	50.4	-0.0826								
84 0	-0 2158		87 0	3 5359	-1 6081	-1.5934	0 8980	0.3264	56.1	0 0041								
87 0	-0 2012		89 0	0 0485	-0 0750	-1 9969	-1 6090	1 0358	62.5	0 0041								
89 0	-0.2125	-0 2125	93 0	0 2953	0.4363	-0 1102	-0 3395	-0 7274	69.7	0.0041								
93 0	-0 2125		96 0	0 2424	-0 5864	1.1945	1.4943	-1 7148	76.9	0.0041								
96 0	-0 2125	-0 1909	100 0	-2 3143	-2 2614	-1 6972	-1 7325	-1.7325	83.4	0.0041								
100 0	-0 2125	-0 2125	96 0	-0 0045	-0 0045	-0.0926	-0 1279	-0.0397	89 4	-0.0176								
96 0	-0.2125	-0 2125	84 0	0.1366	0 1366	0 2248	0.1366	0 0308	95.4	0.0041								
84 0	-0 2125	-0 2125	73 5	-0 1515	-0.0692	0 0131	-0 0692	-0 0692	101 1	-0.0393								
73 0	-0 2125	-0 2125	54 0	-0 0692	-0 0692	-0 0692	-0 0692	-0 1515										
50 0	-0 2125	-0 2125	33 0	-0 0692	-0 0692	-0.0692	-0.0692	-0.1515										
35 0	-0.2172	-0 2172	24 0	0 0044	-0 0241	-0 0526	0.0044	-0.0810										
25 0	-0 1969	-0 2172	10 0	0 0897	0 0897	0 0897	0 1752	0.0954										
10 0	-0 1969	-0 2172	5 0	0.1182	0 2321	0 2321	0.2890	0 2599										
5 0	-0 2172	-0 2172	2 5	0 2605	0.3175	0 3175	0 3744	0 3422										
2 5	-0 2172	-0 2172																

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TABLE A1 (Concluded)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 79				
61 12	CLAERO =	1 0220	CDAERO =	0 2656	DCLC =	0 0	DWLC =	0.4800	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11 99	2112.25	2100 27	11 98	100.59	0.01	2 67	0.0	1.058	1 058	87.2142	0.642746E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	1 50	-0.68	-0 52	0 01	-0 01	0.02	1.04	-0.46	1 01	0.24	

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3 375	BP=10 125
%X/C	CP	CP
0.0	-0 2686	-0.2686
2 5	-0 2686	-0 2686
5.0	-0 2686	-0 2686
10 0	-0 2481	-0 2686
15 0	-0 2686	-0.2686
25 0	-0 2686	-0 2891
35 0	-0.2686	-0 2481
50.0	-0 2481	
56 0	-0 2686	-0.2686
65 0	-0.2891	-0 2891
76 0	-0 2686	-0 2891
79 0	-0 2481	-0 2686
80 5	-0 2854	-0 2558
81 0	-0 2558	-0 2707
82 0	-0 2558	-0 2707
84 0	-0 2558	
87 0	-0 2707	
89 0	-0 2449	-0 2885
93 0	-0 2667	
96 0	-0 2449	-0 2667
100 0	-0 2449	-0 2667
96 0	-0 2667	-0.2667
84 0	-0 2449	-0 2667
73.0	-0 2667	-0 2667
50 0	-0 2667	-0 2449
35 0	-0.2686	-0 2891
25 0	-0 2481	-0 2891
10 0	-0 2686	-0 2686
5.0	-0 2686	-0 2891
2 5	-0.2891	-0 2686

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-2 2452	-4 1417	-5.6645	-3 1072	-1 8429
2 5	-1.4981	-2.2452	-3 8543	-2 3889	-1 6131
5 0	-1 2108	-1 5556	-3.0498	-2 5038	-1 5556
10 0	-0 9809	-1 1821	-2 7337	-2.6188	-1 5556
15 0	-0.8659	-1.0384	-1 3832	-2.2452	-1.4981
24 0	-0 7223	-0 8659	-0 8947	-1 6705	-1 4119
33 0	-0 6597	-0 7428	-0 8259	-0.9089	-1.4904
54 0	-0 5767	-0 6597	-0 6597	0.9187	-1.4074
65 0	-0.6822	-0 7259	-0 7478	-0 6385	-1 5789
78 5	-1 2289	-1.2945	-1.3383	-1.0978	-2 0162
79 5	131.5552	131.5552	131 5552	131.5552	128.3079
80 5	-56 5356	-48 6201	-60 4115	-55 4111	-44.2558
81.3	-23 0846	74.2065	-3 6290	-5.3156	-3 4367
82 0	28 8163	27 7511	20.0726	-4.9014	0 2916
84.0	8.2068	14.1247	3 0286	-7.1503	0 7946
87.0	3.3246	-1 3359	-1.6021	0.6910	-0 5665
89 0	0 0406	-0 1375	-2.1489	-1 6327	0 0049
93 0	0 2541	0 3432	-0 1019	-0.4222	-1.8107
96 0	0 2185	-0.6359	1 1619	1.4646	-2 6651
100 0	-2 2557	-2 2201	-1 7039	-1 7395	-1.8997
96.0	0 0406	0 0406	-0 0307	-0 1731	-0 1553
84 0	0 2007	0 2363	0.3075	0 0939	-0 0484
73 5	-0 0782	0 0879	0 0879	0 0879	-0.0782
54 0	0 0048	0.0048	0 0048	-0 0782	-0 1613
33 0	0.1710	0.1710	0.1710	0.1710	-0.0782
24 0	0 1685	0 2259	0 1972	0.2259	0 0248
10 0	0 3696	0.3409	0 3983	0 4558	0 2541
5 0	0.4845	0 4845	0 5133	0.5133	0.4202
2.5	0 5995	0.5420	0 5133	0.5133	0.5033

\*\*FUSELAGE\*\*

-----TOP----		**MISC.**
XLOC	CP	NO. CP
38.2	-0.1574	1 -0 7428
43.4	-0.1574	2 0.1710
50 4	-0.1792	3 -1.8228
56.1	-0.2230	4 0.0879
62.5	-0.2449	5 -1.2413
69.7	-0.2667	6 -0 1613
76 9	-0.3105	7 0.4499
83.4	-0.2449	8 -0.3510
89.4	-0.2885	9 -0 1375
95.4	-0.2667	10 0.4499
101.1	-0.2230	11 0.0939
---BOTTOM---		12 -0.1553
38 2	-0.0480	
43.4	-0.0043	
50.4	-0.0480	
56 1	0 0613	
62.5	0.1051	
69.7	0 1488	
76 9	0.1270	
83.4	0.1051	
89 4	0.0832	
95.4	0.0832	
101.1	-0 0043	

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TABLE A2 RUN 62, BW6V, DELF=0, CMU=2 O, (BN/B)=1

RUN SEQ				PROPLUSIVE		WING / CANARD		PRESSURES			PAGE 80
62 2	CLAERO =	0.3651	CDAERO =	0 0117	DCLC =	0 0	DWLC =	0 5330	BASEPR =	8 1904	
ALPHA	PTDT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.06	2112 25	2106 37	.5 88	70 40	0 01	2 65	0 0	2.051	2.051	86.7146	0.450351E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	0 90	-1 97	-0 49	0 01	-0 01	0 04	0.32	-0 35	0 32	0.01	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0 0	-0.3325	-0.2908	0.0 0	5354	0 5354	0.1839	0 3011	0 4182	38.2	-0.1551	1 -0 5201	
2 5	-0.3325	-0.3325	2 5	-0 3431	-0 2846	-0 6360	-0 7531	-0 5774	43.4	-0.1551	2 -0.3507	
5 0	-0 3325	-0 3325	5 0	-0 3431	-0 2846	-0 5774	-0.5189	-0 6360	50.4	-0.1551	3 -0.5201	
10.0	-0 3325	-0 3325	10 0	-0 4017	-0 4017	-0.5774	-0 4603	-0.5774	56.1	-0.1551	4 -0 1814	
15 0	-0 3325	-0 3743	15.0	-0.3431	-0 4603	-0.5774	-0 4603	-0 5774	62.5	-0.2888	5 -0 5201	
25 0	-0 3325	-0 3325	24 0	-0 3431	-0 4017	-0 5189	-0 4603	-0 5189	69.7	-0.3334	6 -0 3507	
35 0	-0.3325	-0 3325	33 0	-0 3507	-0 5201	-0.5201	-0 5201	-0.5201	76.9	-0.2442	7 0 4234	
50 0	-0 3325		54 0	-0.5201	-0 5201	-0 6894	1 0039	-0 6894	83 4	-0.1996	8 -0 3386	
56 0	-0 3325	-0 3325	65 0	-0 6900	-0.7791	-0.7791	-0 8237	-0 7791	89.4	-0.2442	9 -0 0846	
65 0	-0.3325	-0.3325	78 5	-1 6260	-1.7152	-1.7597	-1 8043	-1 9826	95.4	-0.3334	10 0 6411	
76 0	-0 3325	-0 3325	79.5267	1089	267 1089	267.1089	267.1089	248.4091	101.1	-0.2442	11 0 2057	
79 0	-0 3325	-0 3325	80 5*****	-86 8113	-114 3440	-104.9345	-74 2658		---	---	12 -0 0483	
80 5	-0 2928	-0 2928	81.3-44	4410	167 0733	-12 2045	-32.5591	0 3706	38.2	-0.1551		
81 0	-0 2325	-0 2325	82 0 59	3262	51 6063	37 6433	23.1683	7.4574	43.4	-0.1105		
82.0	-0 2626	-0 2928	84 0 20	5751	-10 9680	-1 8006	18 3444	1 3658	50 4	-0.1551		
84.0	-0.2626		87.0	6.2813	-3 4593	-1 9212	2.2705	1 3356	56.1	-0.1105		
87.0	-0 2928		89.0	0 6774	0 4597	-3 4225	-2 5880	2.7092	62 5	-0 1105		
89 0	-0.2888	-0 3334	93 0	1.0402	1.2578	0 2420	-0.5199	-0 6288	69.7	-0.1105		
93 0	-0 2888		96 0	0 8225	-0 7738	2 4552	3 3623	-2.6243	76.9	-0.1551		
96 0	-0 3334	-0.2888	100 0	-4 2932	-4 2207	-3 2411	-3 2411	-3 2411	83 4	-0 1551		
100 0	-0 3334	-0 3334	96 0	-0 2297	-0 2297	-0 4111	-0.4836	-0.2297	89 4	-0 1551		
96 0	-0 2888	-0 2888	84 0	-0 0120	0 0243	-0 4474	-0 0483	-0.0483	95.4	-0.1105		
84 0	-0.3334	-0 3334	73 5	-0 3507	-0 1814	-0.1814	-0 1814	-0.1814	101.1	-0 1996		
73 0	-0 2888	-0.2888	54 0	-0 1814	-0 3507	-0 3507	-0 1814	-0 3507				
50 0	-0 2888	-0 2888	33 0	-0 1814	-0 3507	-0 3507	-0 1814	-0 3507				
35 0	-0 3325	-0.3325	24 0	-0 1675	-0 2260	-0 2260	-0 1089	-0 2846				
25 0	-0 3325	-0 3325	10 0	-0 1675	-0 1675	-0 1089	-0 0503	-0 1814				
10 0	-0.3325	-0 3325	5 0	-0 1089	-0 1675	-0 1089	-0 1089	-0 1814				
5 0	-0 3325	-0 3325	2.5	-0 1675	-0 2260	-0 0503	0 0083	-0 0120				
2 5	-0 3325	-0 3325										

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TABLE A2 (Continued)

RUN·SEQ				PROPLULSIVE		WING / CANARD		PRESSURES		PAGE 82	
62 4	CLAERO =	0.5829	CDAERO =	0.0966	DCLC =	0.0	DWLC =	0.6839	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.02	2112.32	2106.44	5.88	70 38	0 01	2.65	0 0	2 099	2.099	86.6010	0.450830E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	1.27	-1.89	-0.55	0.01	-0.01	0 03	0.55	-0 41	0.55	0.05	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3 375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	----	TOP----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	-0 3325	-0 3325	0 0	0 0083	-0.5188	-1.1630	-1.5728	-1.4558	38 2	-0.1551	1	-0 6893
2.5	-0 3325	-0.3325	2 5	-0 5773	-0.9287	-1.6313	-1.6899	-1.6313	43.4	-0.1551	2	-0 0120
5 0	-0.3325	-0.3325	5 0	-0 5188	-0.7530	-0 9287	-1.1043	-1.1043	50.4	-0.1551	3	-0.6893
10.0	-0.3325	-0.3325	10.0	-0 5188	-0 7530	-0.8115	-0.8115	-0.9287	56.1	-0 1551	4	-0.1814
15.0	-0.3325	-0 3325	15.0	-0 5188	-0 6359	-0 8115	-0 7530	-0 8702	62 5	-0.2888	5	-0 6893
25 0	-0.3325	-0 3325	24 0	-0 4602	-0.5773	-0 6944	-0.6359	-0.6359	69.7	-0 2888	6	-0.1814
35.0	-0.3325	-0.3325	33 0	-0 5200	-0 5200	-0 6893	-0 6893	-0 6893	76.9	-0.2441	7	0.4596
50 0	-0 3325	-0.3325	54.0	-0 6893	-0.6893	-0 6893	0 8344	-0.8586	83 4	-0 2441	8	-0 3385
56.0	-0.3325	-0 3325	65.0	-0.7344	-0.8236	-0.8236	-0.9127	-0.9127	89.4	-0.2441	9	-0 1208
65.0	-0.3325	-0.3325	78.5	-1 6703	-1.8040	-1 8486	-1.9377	-2.1606	95.4	-0.3333	10	0.6047
76 0	-0.3325	-0.3325	79.5	267.1191	267.1191	267.1191	267.1191	249.0005	101.1	-0.2888	11	0 2419
79.0	-0.3325	-0 3325	80.5	*****	-88 0331	-115.1397	-105.4016	-75.3704	----	BOTTOM----	12	-0.0483
80.5	-0 2626	-0 2626	81.3	-44.7354	149.8324	-12.1726	-30 6249	-0 2023	38.2	-0.1551		
81.0	-0.2324	-0.2626	82.0	60.1303	52.7740	37.9693	21.7782	7.0944	43 4	-0.1551		
82 0	-0 2626	-0 2626	84.0	23 9490	-8 6749	-1.5290	18.4624	1.4259	50.4	-0.1996		
84.0	-0.2324		87 0	6 3104	-3.4587	-1 8607	2 3304	1 2450	56.1	-0.1104		
87 0	-0.2626		89 0	0 6410	0 3508	-3.5671	-2 5877	2.5636	62.5	-0.0659		
89 0	-0 2888	-0 2888	93 0	1 0400	1.1851	0 2782	-0.4836	-0 8464	69.7	-0.0659		
93.0	-0.2888		96 0	0.7861	-0.8464	2 5636	3 2891	-2 8053	76.9	-0.1104		
96 0	-0 3333	-0 3333	100 0	-4 2563	-4 2200	-3.2044	-3.3131	-3 2768	83.4	-0.1104		
100 0	-0 2888	-0 3333	96 0	-0 1571	-0 1933	-0 3748	-0 4836	-0 1933	89.4	-0.1104		
96 0	-0 2888	-0 3333	84.0	0 0606	0 0606	-0 3385	-0.0483	-0.0483	95.4	-0.1104		
84 0	-0 2888	-0 3333	73 5	-0.3507	-0 0120	-0 1814	-0.1814	-0.1814	101.1	-0.1551		
73 0	-0.2888	-0.3333	54 0	-0 1814	-0 3507	-0.1814	-0 3507	-0.3507				
50 0	-0 3333	-0 3333	33 0	-0 0120	-0 0120	-0 1814	-0.1814	-0.1814				
35 0	-0 3742	-0 3325	24 0	-0 0503	-0 0503	-0 0503	-0.0503	-0.1089				
25.0	-0 3325	-0 3325	10.0	0 1254	0 0668	0.1254	0.1254	-0.0120				
10 0	-0.3325	-0.3325	5 0	0 1254	0.1254	0 2425	0 2425	0 1572				
5.0	-0 3325	-0 3325	2 5	0 1839	0 2425	0.3596	0.4182	0 3265				
2 5	-0 3325	-0 3325										

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TABLE A2 (Concluded)

RUN SEQ	62 8	CLAERO =	1.1089	CDAERO =	0.3144	DCLC =	0 0	DWLC =	0.9205	BASEPR =	8.1904	PAGE	84
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
12 00	2112 32	2106.22	6 10	71 67	0.01	2 67	0.0	2.028	2 028	87.2420	0.460225E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
-0 01	2 03	-1 49	-0.68	0 01	-0 01	0 03	1.12	-0.55	1 10	0.24			

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **	
%X/C	BP=3.375	CP	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	XLOC	CP	NO.	CP
0 0	-0 3679	-0 3276	0 0	-3 3430	-4 8092	-3.1174	-2 6098	-1 8766	38.2	-0.1595	1	-0.7941
2 5	-0 3679	-0 3276	2 5	-1 9332	-4 5273	-3 1174	-2 5534	-1 7076	43 4	-0.1595	2	0.1842
5 0	-0 3276	-0 2874	5 0	-1 4256	-1 5946	-3 2866	-2 3842	-1 7639	50 4	-0.2024	3	-1 9354
10 0	-0 3679	-0 3276	10 0	-1 2000	-1 2000	-3.1738	-2.3278	-1 7076	56 1	-0.2453	4	0 0212
15 0	-0 3276	-0 3679	15 0	-1 0872	-1.0872	-2 2715	-2.3278	-1 6512	62.5	-0.3312	5	-1 6092
25 0	-0 3679	-0 3276	24 0	-0 8616	-0.9744	-1.2563	-2 2151	-1.7076	69 7	-0.2882	6	-0.3050
35 0	-0 3679	-0.3679	33.0	-0 7941	-0 7941	-0 7941	-2 0985	-1 6092	76 9	-0.3741	7	1 0344
50 0	-0 3679		54.0	-0.7941	-0 7941	-0.7941	0 5103	-2.0985	83 4	-0.2882	8	0 1260
56 0	-0 3276	-0 3679	65 0	-0 8462	-0 9750	-0 7604	-0 9321	-1.7475	89.4	-0.3312	9	0.4055
65 0	-0 3679	-0.3679	78 5	-1 7046	-1.8334	-1 7475	-1.4901	-3.1212	95.4	-0.3312	10	1 1042
76.0	-0 3679	-0 3679	79 5257.1965	257.1965	257 1965	257.1965	234.8652	101.1	-0.2453	11	0.6850	
79 0	-0 3679	-0 3276	80.5-93.5741	-80 9142	-107.6284	-98.1617	-69.2127	---	BOTTOM---	12	0.4055	
80 5	-0 3363	-0 3072	81 3-42.2954	150.2793	-11 3993	-37.6194	2.4513	38.2	-0.0736			
81 0	-0 3072	-0 3363	82 0 53 6151	43.2490	34.1017	30.6463	7.8232	43.4	-0.0307			
82 0	-0 3072	-0 3363	84.0 8.2877	-19.4429	-3 6756	18.5959	0 3896	50.4	-0.1166			
84 0	-0 2782		87 0 6.1681	-3 0076	-1.9915	2.1319	-0.4815	56.1	0.0122			
87 0	-0.2782		89 0 0 5103	0 8247	-2.7388	-2 2844	1 8379	62.5	0.0980			
89 0	-0 2453	-0 2882	93.0 0 9645	1 5933	0 8247	-0.5377	-1 4112	69 7	0 0980			
93 0	-0 2882		96 0 1 4536	-0 2932	2.8859	3 1305	-3 4374	76 9	0 0552			
96 0	-0 2882	-0 3312	100 0 -3 0880	-3 2627	-2.3544	-3 1578	-2 4941	83 4	0.0552			
100 0	-0 2882	-0 3312	96 0 0 6850	0 5103	0 3007	-0 0487	0.3356	89.4	0 0122			
96 0	-0 3312	-0 3312	84 0 0 8247	0 7548	-0 1884	0 6151	0 4754	95 4	-0 0307			
84 0	-0 2882	-0 3312	73 5 -0 1419	0 0212	0 0212	0.0212	-0 3050	101.1	-0.0736			
73 0	-0 3312	-0 3312	54 0 0 0212	0 0212	-0 1419	-0 1419	-0.3050					
50 0	-0 2882	-0 3312	33 0 0 1842	0 0212	0 0212	0 0212	-0.1419					
35 0	-0 3679	-0 3276	24 0 0 1535	0 1535	0 0971	0.1535	-0.0721					
25 0	-0 3679	-0 3276	10 0 0 3227	0 3791	0 3227	0 3791	0.1842					
10 0	-0 3276	-0 3679	5 0 0 5483	0 4919	0 4919	0.4919	0 3472					
5 0	-0 3679	-0 3679	2 5 0 6047	0 4919	0 4919	0 4919	0 3472					
2 5	-0 3276	-0 3679										

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TABLE A3. RUN 64, BWGV, DELF=15, CMU=0, (BN/B)=1

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 85				
64 3	CLAERO =	0 4705	CDAERO =	0 1113	DCLC =	0 0	DWLC =	0.0	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.02	2115 72	2085.43	30.29	159.16	0.01	2.67	0.0	0 0	0 0	87.1804	0.103854E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR
-0.01	0.47	0 11	-0.31	0 00	-0 00	0 01	0 47	-0.31		0.47	0.12

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0 0	-0.2152	-0.2152	0 0	0.5345	0.5004	0.4095	0.3413	0.2617	38.2	-0.0721	1 -0.2810	
2.5	-0.2071	-0.2071	2.5	-0.2043	-0.2497	-0.4202	-0.6361	-0.5679	43.4	-0.0549	2 0.0147	
5.0	-0.2152	-0.2071	5.0	-0.2156	-0.2497	-0.3406	-0.4998	-0.4429	50.4	-0.0635	3 -0.3139	
10.0	-0.2071	-0.2071	10.0	-0.2838	-0.3293	-0.3975	-0.4088	-0.4202	56.1	-0.0549	4 0.0147	
15.0	-0.2071	-0.2152	15.0	-0.2724	-0.2952	-0.3634	-0.4656	-0.4088	62.5	-0.1932	5 -0.2481	
25.0	-0.2152	-0.2071	24 0	-0.2611	-0.2838	-0.3406	-0.3747	-0.3634	69.7	-0.1932	6 -0.0510	
35 0	-0.2071	-0.2071	33.0	-0.2153	-0.2481	-0.3139	-0.3139	-0.3139	76.9	-0.1154	7 0.5217	
50.0	-0.2071		54 0	-0.3139	-0.3139	-0.3796	0.1133	-0.3467	83.4	-0.1154	8 -0.2035	
56.0	-0.2071	-0.2071	65.0	-0.4441	-0.4787	-0.4960	-0.5133	-0.4787	89.4	-0.1759	9 -0.0274	
65.0	-0.2071	-0.2071	78.5	-1.1015	-1.1448	-1.1880	-1.2140	-1.2313	95.4	-0.2105	10 0.5288	
76 0	-0.2152	-0.2071	79.5	-1.3988	-1.3988	-1.3637	-1.3988	-1.4456	101.1	-0.1759	11 0.2330	
79.0	-0.2071	-0.2071	80.5	-1.0536	-1.2291	-1.2116	-1.2467	-1.4164	----	----	12 -0.0345	
80.5	-0.2050	-0.1992	81 3	-0.6322	52.6168	-0.8780	-0.9248	-1.1238	38.2	-0.0462		
81 0	-0.1933	-0.1933	82.0	-0.6088	-1.2291	-0.7961	-0.8253	-0.9775	43.4	-0.0116		
82.0	-0.1933	-0.1933	84 0	-0.6205	-0.6849	-0.7083	-0.6966	-0.7785	50.4	-0.0549		
84 0	-0.1933		87.0	-0.5093	-0.5562	-0.5912	-0.5503	-0.6439	56.1	0.0057		
87.0	-0.1933		89.0	-0.4640	-0.4992	-0.5344	-0.5274	-0.5555	62.5	0.0057		
89 0	-0.1846	-0.1932	93.0	-0.3091	-0.3091	-0.3232	-0.2246	-0.5485	69.7	0.0144		
93.0	-0.1932		96.0	-0.2246	-0.2105	-0.2246	-0.2387	-0.4288	76.9	0.0144		
96.0	-0.1932	-0.1932	100 0	-0.1471	-0.0908	-0.1471	-0.1401	-0.2457	83.4	0.0490		
100.0	-0.1932	-0.1846	96.0	0.1415	0.2260	0.2260	0.1979	0.1838	89.4	0.1009		
96 0	-0.2019	-0.1846	84.0	0.3739	0.5076	-0.2457	0.4231	0.5428	95.4	0.1268		
84.0	-0.1932	-0.1846	73.5	0.2776	0.3104	0.3433	0.3433	0.3104	101.1	0.0317		
73 0	-0.2019	-0.1846	54.0	0.0804	0.0476	0.0476	0.0804	0.0147				
50 0	-0.1932	-0.1846	33.0	0.0147	0.0147	0.0147	0.0147	-0.0181				
35 0	-0.2071	-0.2152	24.0	-0.0679	-0.0451	-0.0565	-0.0565	-0.0792				
25.0	-0.2071	-0.2071	10.0	-0.0565	-0.0451	-0.0110	-0.0338	-0.0181				
10.0	-0.2152	-0.2152	5.0	-0.0110	-0.0565	0.0117	0.0230	0.0476				
5 0	-0.2152	-0.2071	2.5	-0.0110	0.0230	0.0458	0.1140	0.2119				
2 5	-0.2152	-0.1990										

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TABLE A3 (Continued)

RUN SEQ 64 5  
 ALPHA 4 04  
 BETA -0 01  
 CLAERD = 0 6538  
 PTOT 2115 93  
 CL 0 65  
 CDAERD = 0.1506  
 VEL 30 40 159 58  
 CM -0 34  
 DCLC = 0 0  
 YAW 0 01  
 H/C 2 66  
 CN -0 00  
 DWLC = 0 0  
 CMUC 0 0  
 CMUW 0 0  
 CMUT 0 0  
 CNTR 0.66  
 CMTR -0.34  
 BASEPR = 8.1904  
 HGT RN  
 87 0821 0.103826E+07  
 CLTR CDTR  
 0.65 0 16

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	----	TOP----	NO	CP
0 0	-0 2030	-0 2030	0.0	0 0262	-0 4040	-0.8909	-1 3212	-1 7402	38.2	-0.0945	1	-0.4404
2 5	-0.1950	-0 2030	2.5	-0 6078	-0.8230	-1 1967	-1 1174	-1 5590	43.4	-0.0686	2	0.0507
5 0	-0 1950	-0 2030	5 0	-0.5059	-0 6305	-0 8230	-0 9136	-0.9589	50.4	-0.0686	3	-0.5386
10.0	-0 1950	-0 1950	10 0	-0.4606	-0.5399	-0.6984	-0 7324	-0 8117	56.1	-0.0772	4	0.0507
15.0	-0 1950	-0 1950	15 0	-0 4380	-0.4946	-0.5965	-0 6418	-0 6418	62.5	-0.1892	5	-0.4731
25.0	-0 1950	-0 1869	24 0	-0.3927	-0 4267	-0 4946	-0 5286	-0 5172	69.7	-0.1806	6	-0.0802
35 0	-0 2030	-0 1950	33 0	-0.3749	-0 4076	-0 4731	-0.5058	-0.5058	76 9	-0.1548	7	0.5231
50.0	-0 2030		54 0	-0.3749	-0 4404	-0 4404	-0.1457	-0 5386	83 4	-0 1375	8	-0.1784
56 0	-0 2030	-0 1950	65 0	-0 4736	-0.5167	-0 5426	-0 5857	-0.6115	89.4	-0.1892	9	-0 0591
65 0	-0 1950	-0 1950	78.5	-1 0769	-1 1286	-1.1631	-1.2234	-1.3613	95.4	-0.2237	10	0 4740
76 0	-0.2111	-0.1950	79.5	-1 3612	-1 3495	-1 3612	-1.3962	-1.5245	101.1	-0.1806	11	0.2705
79 0	-0 1950	-0 1950	80 5	-1 0347	-1.1222	-1 1746	-1 2271	-1.5186	----	BOTTOM----	12	-0 0311
80 5	-0 1951	-0 1835	81 3	-0 6382	46 5919	-0 8481	-0.9298	-1.2971	38 2	-0.0514		
81 0	-0 1835	-0.1893	82 0	-0 6266	-1.1338	-0 7723	-0.8248	-1 1863	43 4	-0.0169		
82 0	-0 1835	-0.1893	84.0	-0 6207	-0.6674	-0 7024	-0.6907	-0 9531	50.4	-0 0427		
84 0	-0 1835		87.0	-0.4867	-0 5391	-0 5741	-0 5566	-0.7898	56 1	0 0176		
87 0	-0.1835		89 0	-0 4379	-0 4519	-0.4870	-0.5011	-0 7045	62 5	0.0607		
89 0	-0.1806	-0.1892	93 0	-0 2906	-0 2836	-0.3117	-0 2205	-0.6764	69.7	0.0779		
93 0	-0.1892		96.0	-0.2064	-0 1924	-0 2205	-0.2626	-0.5431	76.9	0.0693		
96 0	-0 1892	-0.1892	100 0	-0.1363	-0 1152	-0 1784	-0 2556	-0.3467	83.4	0.0952		
100 0	-0 1806	-0 1892	96 0	0.1723	0.2355	0 2144	0.1794	0.1443	89 4	0.1296		
96 0	-0.1892	-0 1892	84 0	0 4249	0 3056	-0 2415	0.4249	0.2215	95.4	0.1469		
84 0	-0.1806	-0 1892	73.5	0.2798	0 2798	0 3453	0.3126	0 2471	101.1	0.0435		
73 0	-0.1892	-0 1892	54 0	0.0834	0 0507	0 0834	0.0834	-0 0475				
50 0	-0.1806	-0 1892	33 0	0.0507	0 0507	0 0507	0 0507	-0 0475				
35 0	-0 2030	-0 1950	24 0	0.0376	0 0489	0 0715	0 0829	0 0149				
25.0	-0 2111	-0 1950	10 0	0.1508	0 1508	0 1848	0.2188	0 1489				
10 0	-0 2030	-0 2030	5 0	0 2188	0 2188	0 2641	0.3094	0 2798				
5 0	-0 2030	-0 2030	2 5	0 3433	0.3207	0 3886	0 4226	0 4108				
2.5	-0 2030	-0 1950										

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TABLE A3. (Concluded)

RUN SEQ				PROPLUSIVE		WING / CANARD		PRESSURES			PAGE 89
64 9	CLAERO =	1 0835	CDAERO =	0 3162	DCLC =	0 0	DWLC =	0.0	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12 03	2116.00	2085 83	30.17	159 09	0 01	2 67	0 0	0 0	0 0	87 3045 0	103263E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	1.08	0 32	-0 44	0.01	-0 00	0.00	1.12	-0 44	1.08	0.32	

***** CANARD *****			***** WING *****					**FUSELAGE**			
	BP=3 375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	**MISC.**	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP
0 0	-0 2600	-0 2437	0 0	-2.8436	-5 5699	-6.1860	-2.8549	-1.6001	38 2	-0 1018	1 -0 7141
2 5	-0 2437	-0 2356	2.5	-1 6001	-2 2845	-3 1856	-2 9119	-1 5772	43 4	-0.0931	2 0.2753
5 0	-0 2437	-0.2437	5.0	-1 1780	-1 6001	-2 3186	-2.7979	-1.6001	50.4	-0.1278	3 -1.5386
10 0	-0 2356	-0 2437	10 0	-0 9384	-1 1666	-1 5316	-3 0375	-1.4632	56.1	-0.1625	4 0.2094
15 0	-0 2519	-0.2437	15.0	-0 8015	-1 0297	-1 1437	-2.4900	-1 5316	62 5	-0.2146	5 -1 3078
25 0	-0 2519	-0 2356	24 0	-0.6760	-0 7673	-0.8700	-1.5088	-1.5202	69.7	-0.2146	6 -0.0215
35 0	-0 2193	-0.2275	33 0	-0.6152	-0.6482	-0.7801	-1.0109	-1 4067	76.9	-0.2667	7 0.5627
50 0	-0 2275		54.0	-0 5162	-0 5822	-0 6482	-0.3513	-1 4727	83 4	-0.2146	8 -0.1581
56 0	-0 2275	-0 2193	65.0	-0.5967	-0.6314	-0.6835	-0.6140	-1.1002	89.4	-0.2494	9 0.0186
65 0	-0 2193	-0 2193	78 5	-1.1523	-1.1697	-1.1697	-1.1957	-1 4736	95.4	-0.2146	10 0.4638
76.0	-0 2275	-0.2193	79.5	-1.2840	-1.3310	-1.3427	-1.4073	-1.5248	101.1	-0 1452	11 0.1811
79 0	-0 2275	-0 2193	80 5	-1 2311	-1.2370	-0 9551	-1 2840	-1 5777	---BOTTOM---		
80.5	-0 2326	-0.2326	81.3	-0 8258	52.8320	-0.7671	-1.0138	-1.4954	38.2	0.0024	12 -0.0379
81 0	-0 2326	-0.2267	82 0	-0 7377	-1.3545	-0.7436	-0 8611	-1 3545	43.4	0.0458	
82 0	-0 2267	-0 2326	84.0	-0 6614	-0.7025	-0 6908	-0.7436	-1.3251	50.4	0.0024	
84.0	-0 2267		87.0	-0.5439	-0.5615	-0.5791	-0.6320	-1.2605	56.1	0.1066	
87 0	-0 2267		89.0	-0 4832	-0.4902	-0.5255	-0.6033	-1 1474	62.5	0.1761	
89 0	-0 2146	-0 2407	93 0	-0 3206	-0 3065	-0 3347	-0 2499	-1 1333	69 7	0.2195	
93 0	-0 2233		96 0	-0 2358	-0 2217	-0.2923	-0.3277	-1.0344	76.9	0.2195	
96 0	-0 2320	-0.2233	100 0	-0 1510	-0 1439	-0.2288	-0 2075	-0.8153	83.4	0.2021	
100 0	-0 2320	-0 2146	96.0	0 1811	0 2730	0 2165	0 1882	-0 0097	89 4	0.2108	
96 0	-0 2320	-0.2146	84.0	0 4568	0 1953	-0.2358	0.4285	0.0610	95 4	0.2021	
84 0	-0 2320	-0 2146	73 5	0 3083	0 3413	0 4072	0 3743	0 2423	101 1	0.0806	
73 0	-0 2320	-0 2146	54.0	0 2094	0 1764	0 2094	0 1764	0 0115			
50 0	-0 2233	-0 2146	33 0	0 2423	0 2423	0 2423	0 2423	0 0774			
35 0	-0 2275	-0 2112	24.0	0 2822	0 2936	0 3164	0 3164	0 1681			
25 0	-0 2275	-0.2193	10.0	0 4762	0 4762	0 4762	0 4762	0 3743			
10 0	-0 2275	-0 2193	5 0	0 5788	0 5332	0 5560	0 5560	0 4402			
5 0	-0 2437	-0 2519	2 5	0 6701	0 5674	0 5104	0 5218	0 4402			
2 5	-0 2437	-0 2275									

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TABLE A4. RUN 65. BWGV, DELF=15,CMU=0 5. (BN/B)=1

RUN SEQ																	PAGE 92
65 4	CLAERO =	0 6761	CDAERO =	0 2135	DCLC =	0 0	DWLC =	0.2475	BASEPR =	8.1904							
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN						
-0 03	2116.78	2088 98	27 80	152 70	0 01	2 67	0 0	0 495	0 495	87.3269	0 990272E+06						
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR							
-0 01	0 92	-0 22	-0 61	0 01	-0 00	0 02	0 69	-0.92	0 69	0.14							

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **			
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	XLOC	CP	NO.	CP
0 0	-0 2802	-0 2714	0 0	0 4924	0 3809	0 0961	0 1457	-0.1639	38.2	-0 1026	1	-0 4319	
2 5	-0 2714	-0 2714	2 5	-0 2753	-0 3744	-0 6468	-0 7582	-0 8696	43 4	-0.0837	2	-0 0023	
5 0	-0 2714	-0 2714	5 0	-0.3000	-0 3248	-0 5229	-0 6220	-0 7954	50 4	-0 0837	3	-0.5035	
10 0	-0 2714	-0 2714	10 0	-0 3248	-0.3991	-0 5353	-0 5972	-0 6096	56 1	-0.0931	4	-0 0023	
15 0	-0 2802	-0.2626	15 0	-0 3125	-0 3991	-0 4858	-0.5229	-0 5477	62 5	-0 2627	5	-0.4319	
25 0	-0 2802	-0 2626	24 0	-0 3248	-0 3867	-0 4239	-0.4239	-0 4858	69.7	-0.2627	6	-0 1097	
35 0	-0 2714	-0 2714	33 0	-0.3603	-0.3961	-0 4319	-0 5035	-0 5035	76 9	-0.1780	7	0 4784	
50 0	-0 2714		54 0	-0 4676	-0 5035	-0.5751	-0 3245	-0.5751	83.4	-0.1780	8	-0.4650	
56 0	-0 2802	-0 2802	65.0	-0 6774	-0 7340	-0 7528	-0 7999	-0 7811	89.4	-0.2627	9	-0.1735	
65 0	-0 2714	-0 2714	78.5	-1.6386	-1 7517	-1 8271	-1.9025	-1.9684	95.4	-0.3004	10	0 7009	
76 0	-0 2714	-0 2714	79 5	-5.2477	-12.3307	-23 2325	-18 8719	-20 9184	101.1	-0.2345	11	0.2177	
79 0	-0 2714	-0 2714	80 5	-8.3845	-3.1757	0.3498	5 3416	-1.6075	---	BOTTOM---	12	-0.0662	
80 5	-0 2814	-0.2686	81 3	3.6904	42 1274	-2.1494	-6 2042	-5.9617	38 2	-0.0649			
81 0	-0 2623	-0 2686	82 0	-3 7688	-2 4490	1 7970	-0.8424	-3.3098	43.4	-0.0272			
82 0	-0.2623	-0 2623	84 0	-1 2568	-1 0337	-1 7414	-2 0474	-1 2058	50.4	-0.0743			
84 0	-0.2623		87 0	0 4772	-1.6903	-1 5182	-0 5173	-0 9126	56.1	0.0011			
87 0	-0 2559		89 0	-0 7028	-0 7718	-1 6923	-1 5389	-0.3039	62.5	0.0200			
89 0	-0 2439	-0 2627	93 0	-0.4190	-0.3423	-0 6107	-0.3576	-1.1093	69.7	0.0294			
93 0	-0 2627		96.0	-0 3576	-0 6644	0 0489	0 1716	-1 3701	76 9	0.0388			
96 0	-0 2627	-0 2627	100.0	-0.9559	-0 8025	-0.6951	-0.7028	-0 7872	83.4	0.0765			
100 0	-0 2627	-0 2627	96.0	0 1793	0 2713	0 3020	0.2177	0 1716	89.4	0.1236			
96 0	-0 2627	-0 2627	84 0	0 4094	0 2253	0 1716	0 4478	0 2790	95 4	0.1424			
84 0	-0 2627	-0 2627	73 5	0 2125	0.3199	0.3557	0 3557	0 2125	101.1	0.0388			
73 0	-0 2627	-0 2627	54 0	0 0693	0 0693	0 0693	0 0693	-0.0381					
50 0	-0.2533	-0 2627	33 0	-0 0023	-0 0023	-0 0023	0.0335	-0.0739					
35 0	-0 2714	-0 2714	24 0	-0.0029	0 0095	0 0342	0.0219	-0 0153					
25 0	-0 2714	-0 2714	10 0	0 0219	0 0342	0 0714	0.1209	0.0335					
10 0	-0 2626	-0 2714	5 0	0 0714	0 0590	0 1085	0 1457	0 1409					
5 0	-0.2714	-0 2714	2 5	0 0838	0.1457	0 1085	0 2448	0 2483					
2 5	-0 2714	-0 2714											

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TABLE A4 (Continued)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 94				
65 6	CLAERO =	0 7718	CDAERO =	0 2694	DCLC =	0 0	DWLC =	0.2816	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4 04	2116.92	2088 89	28 03	153 32	0 01	2.65	0 0	0 503	0.503	86.8381	0.994294E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	1.05	-0.15	-0 62	0.01	-0.01	0.03	0.80	-0.93	0.79	0.19	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3 375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0 0	-0.2796	-0 2796	0.0	-0.1186	-0 7449	-2.0469	-2 0960	-3.1276	38.2	-0.1226	1 -0.5988	
2 5	-0 2708	-0 2708	2.5	-0.7449	-0.9537	-1 4082	-1 5433	-1.7644	43.4	-0.0946	2 0 0759	
5 0	-0 2796	-0.2708	5.0	-0.5853	-0 7327	-0 9906	-1.2485	-1 3713	50.4	-0.1039	3 -0 7053	
10 0	-0.2708	-0.2796	10.0	-0.5484	-0.7081	-0 9169	-0.9414	-0.9906	56.1	-0.1133	4 0 0759	
15 0	-0 2708	-0 2796	15.0	-0.5116	-0 6221	-0.7572	-0.8186	-0.8678	62.5	-0.2628	5 -0.6698	
25 0	-0 2708	-0 2708	24.0	-0 4748	-0 5362	-0 6344	-0.6958	-0 6958	69.7	-0.2722	6 -0.1017	
35 0	-0 2796	-0 2708	33.0	-0 4923	-0.5633	-0 6343	-0.6698	-0.6698	76.9	-0.2348	7 0 4665	
50 0	-0 2796		54.0	-0 5633	-0.5988	-0 7053	-0 2437	-0.7763	83.4	-0.2255	8 -0.4998	
56 0	-0.2796	-0 2884	65.0	-0.7302	-0 8050	-0 8424	-0 8985	-0 9452	89.4	-0.3189	9 -0.2182	
65 0	-0.2796	-0.2884	78 5	-1 6744	-1.7866	-1.9081	-2 0202	-2.2166	95.4	-0.3189	10 0 6567	
76 0	-0 2796	-0 2796	79.5	-4 9517	-11 9964	-22.8928	-18.4785	-20.6098	101.1	-0.2722	11 0.2078	
79 0	-0 2796	-0 2796	80.5	-8 9927	-3 9336	-0.4301	4.7618	-1.1447	---BOTTOM---		12 -0 1117	
80 5	-0 2720	-0.2467	81 3	3 9081	56.8064	-2 2451	-6.8300	-6 7415	38.2	-0.0479		
81 0	-0.2593	-0.2467	82.0	-4 0728	-2.8774	2.1627	0.1517	-3.0041	43.4	-0.0105		
82 0	-0 2657	-0.2593	84 0	-1 3660	-0 9993	-2.1376	-1.3281	-1.3534	50.4	-0.0572		
84 0	-0 2657		87 0	0.4995	-1.7265	-1.5494	-0.5692	-1 0056	56.1	0.0269		
87 0	-0 2530		89.0	-0 6976	-0 7813	-1 6106	-1 5954	-0.5150	62.5	0.0643		
89 0	-0 2441	-0.2628	93.0	-0 4237	-0 3552	-0.6443	-0.3704	-1.4052	69.7	0.0924		
93 0	-0 2628		96.0	-0 3628	-0.6748	-0.0052	0.0861	-1.6106	76.9	0.1111		
96 0	-0.2535	-0.2628	100.0	-0 9259	-0.7737	-0 6976	-0.6824	-0 8650	83.4	0.1297		
100 0	-0.2535	-0 2722	96 0	0.2231	0.2839	0.2611	0 2231	0 1241	89.4	0.1671		
96.0	-0 2628	-0 2628	84 0	0.4589	0.2611	0.1546	0.4437	0 2383	95.4	0.1858		
84 0	-0.2535	-0.2722	73 5	0.2179	0.3600	0.3600	0 3600	0.2179	101.1	0.0643		
73 0	-0.2628	-0 2628	54.0	0 1114	0 1114	0 1114	0 1114	-0 0307				
50.0	-0.2628	-0.2535	33.0	0 0759	0 0759	0.0759	0 1114	-0.0307				
35.0	-0.2708	-0.2796	24.0	0 0780	0 0902	0.1148	0.1271	0.0534				
25.0	-0.2796	-0.2708	10 0	0.1762	0.1885	0.2499	0 2622	0 2179				
10 0	-0.2796	-0.2796	5.0	0 2499	0 2990	0.3359	0.3481	0.3245				
5 0	-0 2708	-0.2796	2.5	0.3481	0 3727	0 4587	0 4956	0.4310				
2 5	-0 2708	-0 2796										

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TABLE A4 (Concluded)

RUN SEQ	PROPLULSIVE	WING / CANARD	PRESSURES	PAGE	96
65 10	CLAERO = 1 0131 CDAERO = 0 4264 DCLC = 0 0 DWLC = 0.3421		BASEPR = 8 1904		
ALPHA	PTOT PSTAT 0 VEL YAW H/C CMUC CMUW CMUT		HGT RN		
12 00	2117 34 2089.77 27 57 152 03 0.01 2.67 0.0 0.511 0 511		87.3361 0 986863E+06		
BETA	CL CD CM CROLL CN CY CNTR CMTR		CLTR CDTR		
-0 01	1 36 0 05 -0 65 0 01 -0 01 0 06 1 09 -0.97		1.04 0.35		

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **	
	BP=3 375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO CP
0 0	-0 3201	-0 3023	0 0	-3 2469	-6 3052	-7.7907	-3 7586	-2 2482	38.2	-0.1238	1 -0.8793
2 5	-0.3112	-0 3112	2 5	-1 8363	-2 5229	-4 4828	-3.9457	-2 1234	43.4	-0.1238	2 0 3116
5 0	-0.3112	-0 3112	5.0	-1 3620	-1 8238	-3.1718	-4 1956	-2 2233	50.4	-0.1523	3 -2.2868
10 0	-0 3112	-0 3112	10 0	-1 1248	-1 4369	-1.6740	-4 0085	-2 1858	56.1	-0.2093	4 0.2394
15 0	-0 3112	-0 3112	15.0	-1 0000	-1.1997	-1 5866	-1 8738	-2.1609	62.5	-0.2948	5 -1.8538
25 0	-0 3112	-0.3201	24.0	-0 8751	-0 9625	-1.1872	-0 9001	-2 1109	69.7	-0 2948	6 -0.0493
35 0	-0 3112	-0 3023	33 0	-0 7350	-0 8433	-0 9876	-0.8072	-2 2507	76.9	-0.3613	7 0.4869
50.0	-0 3112		54 0	-0 6989	-0 7711	-0.9154	-0 3380	-2 3230	83.4	-0.3043	8 -0.5338
56 0	-0 3112	-0 3023	65 0	-0 8838	-0 9503	-1 0358	-0.9693	-1.9954	89.4	-0.3613	9 -0 1472
65 0	-0.3023	-0 3023	78 5	-1 8054	-1.9479	-1 8719	-2 2424	-2.4134	95.4	-0.3138	10 0 6880
76.0	-0.3112	-0 2934	79 5	-4 8451	-12 0440	-23.1566	-18 7220	-20 9843	101.1	-0.2378	11 0.0616
79 0	-0 3112	-0 3023	80.5	-10 0192	-4 8836	-0 6352	3 8061	-0 9438	---BOTTOM---		12 -0.1395
80 5	-0 3139	-0 2946	81 3	4 0310	57.7065	-1 9657	-7.6989	-7 5767	38.2	-0.0003	
81 0	-0 3074	-0 2946	82 0	-4 4916	-3 3477	2 3085	0 9973	-2 8142	43.4	0 0377	
82 0	-0 3074	-0 2882	84 0	-2 0621	-0 9823	-2 0364	-0 5581	-2 1778	50.4	-0.0193	
84 0	-0 3010		87 0	0 4574	-1 8693	-1 5929	-0.8024	-2 0428	56.1	0.1137	
87 0	-0 2946		89 0	-0.7581	-0 8277	-1 6706	-1 7324	-1 4540	62.5	0 1897	
89 0	-0 2663	-0 2948	93.0	-0 4642	-0 4023	-0 7504	-0.3946	-2 5057	69.7	0 2468	
93 0	-0 2948		96.0	-0.4023	-0 7039	-0.0466	-0 0234	-2.6449	76.9	0.2468	
96 0	-0 2948	-0 2948	100.0	-0 8586	-0 6962	-0 6498	-0.6653	-0.9050	83.4	0.2563	
100 0	-0 2948	-0 2948	96 0	0 2395	0 3709	0 3091	0.2627	0 0848	89.4	0.2563	
96 0	-0 2948	-0 2948	84 0	0.5024	0 2395	0 1390	0.4792	0.1158	95.4	0.2658	
84 0	-0 2948	-0 2948	73 5	0 3116	0 4199	0 4560	0 3838	0.2033	101.1	0.1137	
73 0	-0 2948	-0 2948	54 0	0 2394	0 2394	0.2394	0.2394	0 0229			
50 0	-0 2948	-0 2948	33 0	0.2755	0.2755	0.3116	0.3116	0 0951			
35 0	-0 3112	-0 3112	24 0	0.2983	0 3232	0 3482	0.3357	0 1859			
25 0	-0.3023	-0 3112	10 0	0 4855	0 4855	0 4980	0 5105	0 3838			
10 0	-0.3023	-0 3112	5.0	0 5854	0 5729	0 5479	0.5604	0 4560			
5 0	-0.3112	-0 3201	2 5	0 6728	0 5479	0 4605	0.4605	0 3838			
2 5	-0 3023	-0 3112									

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RUN SEQ	66 3	CLAERO =	0 7803	CDAERO =	0.3068	DCLC =	0 0	DWLC =	0.5284	BASEPR =	8.1904
ALPHA	0 09	PTOT	2117 77	PSTAT	0 13 34	VEL	105.36	YAW	0 01 2 66	CMUC	0 0
BETA	-0.01	CL	1.31	CD	-0 60	CM	-0.85	CROLL	0.00	CN	-0.00
										CY	0 01
										CNTR	0.81
										CMTR	-1.50
										CLTR	0.81
										CDTR	0.15

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10.125
%X/C	CP	CP
0 0	-0.3756	-0 3572
2 5	-0 3756	-0 3388
5 0	-0 3572	-0 3572
10 0	-0 3572	-0 3572
15 0	-0 3572	-0 3572
25.0	-0.3572	-0.3572
35 0	-0.3572	-0 3572
50.0	-0.3572	-0 3572
56.0	-0.3572	-0.3572
65 0	-0.3756	-0 3756
76 0	-0.3756	-0 3756
79.0	-0.3756	-0 3572
80 5	-0 3507	-0 3242
81 0	-0 3375	-0 2976
82 0	-0.3507	-0.2976
84 0	-0 3375	
87.0	-0 3242	
89 0	-0 3114	-0 3310
93 0	-0 3310	
96 0	-0 3310	-0 3310
100 0	-0 3310	-0.3310
96 0	-0 3310	-0 3310
84 0	-0.3310	-0.3310
73 0	-0 3310	-0 3310
50 0	-0 3310	-0.3310
35.0	-0 3756	-0 3572
25 0	-0 3756	-0 3756
10 0	-0.3756	-0.3756
5 0	-0 3572	-0 3756
2 5	-0 3572	-0 3572

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0 0	0.4112	0.3338	0.1273	-0.1308	-0 2599
2.5	-0.3632	-0 4664	-0.6987	-0.9052	-1 0859
5 0	-0.3374	-0.4406	-0 5955	-0.7503	-0 8794
10 0	-0 4148	-0 4664	-0 6213	-0.6729	-0 7245
15.0	-0.4406	-0 4922	-0.6213	-0.6213	-0.6213
24.0	-0.4148	-0.4922	-0 5696	-0.5696	-0.5955
33 0	-0 4131	-0 4131	-0.5623	-0.5623	-0.5623
54.0	-0 5623	-0.5623	-0 7115	-1.4577	-0.7115
65.0	-0.8222	-0.9007	-0 9204	-0.9793	-0.9400
78.5	-2 0991	-2 2367	-2.3348	-2.4134	-2.5510
79.5	-10.5977	-25 5890	-48 6217	-39.3446	-43.0798
80 5	-12.1125	-1.4007	3 1048	14.7072	-3 9392
81 3	8 4209	118.3052	1.4833	-7.2088	-8.0060
82.0	-5 4410	-4 8296	3 0117	-6.2517	-4.0455
84 0	0 6726	-0.8292	-0.2577	-5.1884	-0.9355
87 0	1 9617	-2.5836	-1 9855	0 1809	-0.4571
89.0	-0 6421	-0 6901	-2.2731	-1.7614	0.4772
93.0	-0.3063	-0.1144	-0 5941	-0.3063	-1.2497
96.0	-0 2583	-0.8979	0.9249	0.9729	-1.8893
100 0	-1 8094	-1 6335	-0 7221	-1 2337	-1.3457
96 0	0.1255	0 2054	0 5731	0.1894	0 0775
84 0	0.3973	-0.4182	0.9889	0 5092	-0.3702
73.5	0.1093	0 3332	0.3332	0.3332	0.1840
54.0	0 0347	0 0347	0.0347	0.0347	-0.0399
33 0	-0 0399	-0 0399	-0 0399	-0 0399	-0.1145
24 0	-0 0793	-0.0534	-0 0534	-0.0534	-0.1308
10.0	-0.0276	-0 0276	0 0240	0.0757	-0.0399
5 0	-0.0276	-0 0276	0.0757	0.1015	0 0347
2.5	-0 0018	0.0240	0 1273	0 2047	0.1840

\*\*\*FUSELAGE\*\*

----	TOP----
XLOC	CP
38.2	-0.1542
43.4	-0.1149
50.4	-0.1149
56.1	-0.1345
62.5	-0.3703
69.7	-0.3507
76.9	-0.2524
83.4	-0.2524
89.4	-0.3310
95.4	-0.3507
101.1	-0.2917
---	BOTTOM---
38.2	-0.1149
43.4	-0.0559
50.4	-0.1345
56.1	-0.0364
62.5	-0.0364
69.7	-0.0167
76.9	0.0029
83.4	0.0422
89.4	0.0815
95.4	0.1012
101.1	-0.0364

\*\*MISC \*\*

NO.	CP
1	-0.4877
2	-0 0399
3	-0 5623
4	-0 0399
5	-0 5623
6	-0 1145
7	0.4292
8	-0.5782
9	-0.2424
10	0.7330
11	0.1255
12	-0.1464

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TABLE A5 (Continued)

RUN SEQ	66 5	CLAERO =	0 9923	CDAERO =	0 3862	DCLC =	0 0	DWLC =	0 5921	BASEPR =	8.1904	PAGE	104
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
4 02	2118 12	2104 67	13.45	105 76	0 01	2.66	0 0	1.058	1 058	87.1585	0.693369E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
-0 01	1 58	-0.49	-0.91	0.00	-0 01	0 02	1.05	-1 56	1.03	0.22			

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	----TOP----	NO	CP	
0 0	-0.3724	-0 3541	0 0	-0 1297	-0.5136	-1 9981	-2.2284	-3.1242	38.2	-0 1334	1 -0.6315	
2 5	-0.3724	-0 3724	2 5	-0 7951	-1 0255	-1.3327	-1.4862	-1.9725	43.4	-0.1139	2 0.0344	
5 0	-0 3724	-0 3541	5 0	-0.6160	-0 7951	-1 0511	-1 2303	-1 4094	50.4	-0.1334	3 -0.7795	
10 0	-0 3359	-0 3724	10 0	-0 6160	-0 7440	-0 9488	-0 9999	-1 0767	56.1	-0.1723	4 0 0344	
15 0	-0.3541	-0 3724	15 0	-0 5648	-0 6672	-0 8208	-0 9231	-0 9231	62.5	-0.3672	5 -0 7795	
25 0	-0.3541	-0 3541	24 0	-0 5136	-0.5904	-0 6928	-0 7696	-0 8208	69.7	-0 3477	6 -0 1135	
35 0	-0 3541	-0 3359	33 0	-0 5575	-0 5575	-0 7055	-0 7795	-0 7795	76.9	-0.3087	7 0 4414	
50 0	-0 3359		54 0	-0 6315	-0 7055	-0.8534	-1.2974	-0 8534	83.4	-0.2892	8 -0 6208	
56 0	-0.3541	-0 3359	65 0	-0 8931	-0 9710	-1 0100	-1.0879	-1 1463	89.4	-0 3282	9 -0 2086	
65 0	-0 3541	-0 3541	78.5	-2.1203	-2 2956	-2 4125	-2.5488	-2 8410	95.4	-0.3672	10 0 7427	
76 0	-0.3541	-0 3359	79.5-10	48 18	-25 2673	-48 1701	-38 9590	-43.0307	101.1	-0.3087	11 0 1402	
79 0	-0.3541	-0 3359	80.5-12	7.087	-2 0345	2.7095	14.1877	-3.9190	---	BOTTOM---	12 -0 1451	
80 5	-0 3609	-0 3346	81.3	8.4683	117.3322	1 0622	-8.1225	-8 7813	38.2	-0.0944		
81 0	-0 3346	-0.3346	82.0	-5 6188	-4.3011	2.9599	-5.6980	-4.6174	43.4	-0.0361		
82 0	-0 3346	-0.3214	84.0	0 5220	-0 7958	-0 3478	-5.4871	-1 3756	50.4	-0.1139		
84.0	-0 3478		87 0	1 9056	-2.5617	-2 0609	-0 0579	-0 9276	56.1	-0.0166		
87 0	-0 3478		89.0	-0.6842	-0 7318	-2.4758	-2.1112	0.0768	62.5	0.0224		
89 0	-0.3282	-0 3477	93.0	-0.3196	-0 1611	-0 6525	-0.4464	-1 5879	69.7	0.0613		
93 0	-0 3477		96.0	-0.3037	-0 9379	0 4573	0.7903	-2 1904	76 9	0 0613		
96 0	-0 3477	-0 3477	100 0	-1 7624	-1 6355	-1 3818	-1.3184	-1 5087	83.4	0 1003		
100 0	-0 3477	-0 3282	96.0	0 1561	0 2512	0 1561	0 1085	0 0451	89.4	0.1393		
96 0	-0 3477	-0.3477	84 0	0 4256	-0 3354	0 6793	0.4256	-0.2403	95 4	0.1393		
84 0	-0 3477	-0 3282	73.5	0 1084	0 3304	0 3304	0 3304	0 1084	101.1	0.0029		
73 0	-0 3477	-0 3477	54 0	0 1084	0 0344	0 0344	0 1084	-0 1135				
50 0	-0 3477	-0 3477	33 0	0 0344	0 0344	0 0344	0 0344	-0 0396				
35 0	-0 3359	-0.3541	24 0	0 0238	0 0750	0.1006	0 1006	0 0238				
25 0	-0 3541	-0 3724	10 0	0.1518	0 1773	0 2030	0.2542	0 1824				
10 0	-0 3541	-0 3541	5 0	0 2286	0 2542	0 3053	0.3565	0 3304				
5 0	-0.3541	-0 3541	2 5	0 3053	0 3821	0 4077	0.4589	0 4044				
2 5	-0 3541	-0 3359										

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TABLE A5. (Concluded)

RUN SEQ	66 10	CLAERO =	1.5637	CDAERO =	0 6391	DCLC =	0.0	DWLC =	0.7155	BASEPR =	8.1904	PAGE 106									
ALPHA	11.95	PTOT	2118 26	PSTAT	Q 13.22	VEL	104 78	YAW	0 01	H/C	2.66	CMUC	0.0	CMUT	1.070	HGT	87.1563	RN	0.688724E+06		
BETA	-0.01	CL	2.28	CD	-0.16	CM	-1 11	CROLL	0 00	CN	-0.01	CY	0 02	CNTR	1.69	CMTR	-1 77	CLTR	1.62	CDTR	0.49

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10.125	BP = 2	BP = 6	BP =12	BP =16	BP = 22	----	TOP----	NO	CP
%X/C	CP	CP	%X/C	CP	CP	CP	CP	XLOC	CP		
0 0	-0 4049	-0 3678	0 0	-3 5236	-5 9704	-4 7991	-3.2372	-2 2481	38.2	-0.1829	1 -0.8757
2 5	-0.3864	-0 3678	2 5	-1 9356	-2.6385	-4.6168	-3 0290	-2 1699	43.4	-0.1829	2 0 3285
5 0	-0 3864	-0 3864	5 0	-1 4671	-1 9617	-4.0442	-3.1591	-2 2481	50.4	-0.2027	3 -2.6066
10 0	-0.3864	-0 3864	10 0	-1 2849	-1 4932	-2.8207	-3.3673	-2.1960	56.1	-0.2621	4 0.2532
15 0	-0 3864	-0 3864	15 0	-1 0766	-1.2849	-2 4563	-3.4454	-2 2481	62.5	-0.3612	5 -2.0798
25 0	-0 3864	-0 3678	24 0	-0 8944	-1.0246	-1.3369	-2.4563	-2.1960	69.7	-0.3612	6 -0 0478
35 0	-0 3678	-0 3678	33 0	-0 8004	-0.8757	-0.8757	-1.4777	-2 0798	76.9	-0.4206	7 0 3769
50 0	-0.3678		54 0	-0 8004	-0 8004	-0 9509	-1.2519	-2 7571	83.4	-0 3612	8 -0 7680
56 0	-0.3678	-0 3864	65 0	-1.0546	-1 1537	-1 1537	-1.1140	-2.9368	89.4	-0.4206	9 -0 3003
65 0	-0 3864	-0 3864	78 5	-2.3027	-2.4613	-2 5801	-2 6198	-3.4122	95.4	-0.3612	10 0 7478
76 0	-0.3864	-0 3678	79 5-10	4001	-25.4246	-48.8134	-39 3504	-43 8274	101.1	-0.3017	11 -0 0262
79 0	-0 3864	-0.3864	80 5-13	8448	-2.7202	2 6141	14 4755	-3.9265	---	BOTTOM---	12 -0.2359
80 5	-0 3881	-0 3613	81.3	8 6992	113.3101	0 4831	-9.3012	-9.5289	38.2	-0.0442	
81 0	-0.3747	-0.3479	82 0	-6.1112	-4 2347	3 0967	-4 1677	-4 5430	43.4	-0.0046	
82 0	-0 3479	-0 3613	84 0	0 1078	-0.7231	-0 7634	-4.8111	-2 3583	50.4	-0.0640	
84 0	-0.3613		87 0	1 8904	-2 6666	-2 1841	-0 0530	-2.4522	56 1	0 0746	
87 0	-0 3613		89 0	-0 7358	-0.8486	-2.6224	-2 2354	-1.4291	62 5	0.1539	
89 0	-0.3414	-0 3612	93 0	-0.3809	-0 2359	-0.7358	-0.4938	-3.3481	69.7	0.1935	
93 0	-0 3612		96 0	-0 3326	-0.9937	0 3124	0 7317	-3 7835	76.9	0.2133	
96 0	-0.3810	-0 3612	100 0	-1.7032	-1 5581	-1 3324	-1 3647	-1 5581	83.4	0.2332	
100 0	-0 3612	-0 3612	96 0	0 2157	0.3124	0 2157	0.1190	-0.0262	89.4	0.2332	
96 0	-0 3612	-0 3612	84 0	0 4737	-0.1069	0.6349	0.4414	-0 1230	95.4	0.2133	
84 0	-0 3612	-0 3612	73 5	0 3285	0 4790	0.4790	0 4790	0 1779	101.1	0 0548	
73 0	-0 3612	-0 3612	54 0	0 2532	0 3285	0.3285	0.2532	0 0275			
50 0	-0 3612	-0 3612	33 0	0 3285	0 3285	0 3285	0.3285	0 0275			
35 0	-0 3864	-0 3678	24 0	0 2770	0 3290	0 3290	0.3550	0 1468			
25 0	-0 3864	-0 3678	10 0	0 4331	0 4852	0 5112	0.5112	0 4037			
10 0	-0 3678	-0 3678	5 0	0 5633	0 5633	0 5373	0 5633	0.5543			
5 0	-0 3864	-0 3864	2 5	0 6674	0.5373	0 4592	0.4592	0 4037			
2 5	-0 3864	-0 3678									

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TABLE A6. RUN 67, BW6V, DELF=15, CMU=2.0, (BN/B)=1

RUN SEQ 67 2      CLAERO = 0 8491    CDAERO = 0 4339    DCLC = 0 0      DWLC = 1 0115      BASEPR = 8.1904      PAGE 107  
 ALPHA PTOT    PSTAT    0    VEL    YAW    H/C    CMUC    CMUW    CMUT    HGT    RN  
 0 03    2118 76    2111 75    7 01    76 09    0 01    2 66    0 0    2.021    2.021    87.1544    0.503651E+06  
 BETA    CL    CD    CM    CROLL    CN    CY    CNTR    CMTR    CLTR    CDTR  
 -0 01    1 86    -1.32    -1.19    0 00    -0 01    0 03    0.90    -2 43    0 90    0.13

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**			
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP = 12 CP	BP = 16 CP	BP = 22 CP	XLOC	CP	NO.	CP	
0 0	-0 4371	-0 4371	0 0	0 4586	0.3603	-0 1799	-0 2291	-0 7202	38.2	-0.1974	1	-0.5632	
2 5	-0 4371	-0 4371	2 5	-0 3764	-0 4746	-0 8676	-1 0149	-1.2605	43.4	-0 1974	2	-0.1372	
5 0	-0.4021	-0 4371	5 0	-0 4255	-0 4746	-0 7693	-0.9167	-0 9658	50 4	-0 1974	3	-0.8472	
10 0	-0 4021	-0 4371	10 0	-0 4746	-0 5237	-0 7202	-0 8185	-0 8676	56.1	-0.1974	4	-0 1372	
15 0	-0 4371	-0 4371	15 0	-0 4746	-0 5237	-0 6711	-0.7693	-0.7693	62.5	-0 4591	5	-0.7052	
25 0	-0.4371	-0.4371	24 0	-0.4746	-0.4746	-0 6220	-0 7202	-0.7202	69 7	-0 4217	6	-0.2792	
35.0	-0 4371	-0 4371	33 0	-0.5632	-0 5632	-0 7052	-0 7052	-0 8472	76.9	-0.3095	7	0.3295	
50 0	-0.4371		54.0	-0 8472	-0 8472	-0 9892	-3.5452	-0.9892	83 4	-0.3095	8	-0.7964	
56 0	-0 4371	-0 4371	65 0	-0 9825	-1 1693	-1.1693	-1 2067	-1 2067	89.4	-0.3843	9	-0.3704	
65 0	-0 4021	-0 4371	78 5	-2.7020	-2 8889	-2.9637	-3.1506	-3 3750	95 4	-0.4591	10	0.6946	
76 0	-0 4371	-0 4371	79 5	-20 5228	-48 6715	-92 8023	-75 0478	-81.8773	101 1	-0.3469	11	-0 0661	
79 0	-0 4371	-0 4722	80 5	-18 5508	2 9463	8 3583	30 5131	-7 3975	---BOTTOM---			12	-0.2791
80 5	-0 4174	-0 4174	81 3	16 5273	224 0789	7 7766	-6.7147	-12.3037	38.2	-0.1226			
81 0	-0 4174	-0 3921	82 0	-8.3584	-5 5260	5 9305	-16.0978	-5 5513	43 4	-0.0852			
82 0	-0 3921	-0 4174	84 0	2 7439	-0 9991	0 9736	-8 0550	-0 5438	50 4	-0 1974			
84 0	-0 4174		87 0	4 6154	-3 8569	-2 7694	1.0747	-0 2909	56 1	-0.0852			
87 0	-0 4174		89 0	-0 5530	-0 5530	-3.6565	-2.8958	1 5161	62 5	-0 0852			
89 0	-0 3843	-0 4217	93 0	-0 0661	0 2686	-0.4313	-0 6442	-1.7397	69 7	-0.0478			
93 0	-0 4591		96 0	-0 1270	-1 3441	1.3336	2.1247	-3.1392	76 9	-0.0478			
96.0	-0 4217	-0 4591	100.0	-3 2306	-3.0784	-2 5002	-2.5916	-2 5611	83.4	-0.0478			
100 0	-0 4217	-0.4217	96 0	-0 0053	0.0556	-0 0661	-0.1878	-0 1270	89 4	0.0269			
96 0	-0 4217	-0 4217	84 0	0 2991	-1 1310	1 6987	0 2686	-0 7355	95 4	0.0269			
84 0	-0 4217	-0 4217	73 5	0 1468	0 2888	0 2888	0 2888	0 1468	101 1	-0 1226			
73 0	-0 4217	-0 4217	54 0	0 0048	0 0048	0 0048	0 0048	-0 2792					
50 0	-0 4217	-0 4217	33 0	0 0048	-0.1372	-0 1372	0 0048	-0 2792					
35 0	-0 4371	-0.4371	24.0	-0.0817	-0 0817	-0 0817	-0 0817	-0.1308					
25 0	-0 4371	-0 4371	10 0	-0 0326	-0 0817	-0 0817	0.0656	0 0048					
10 0	-0 4371	-0 4371	5.0	-0 0326	-0 0326	0 0165	0.1639	0 0048					
5 0	-0 4371	-0 4371	2 5	-0 0326	0.1148	0 2130	0.2130	0.1468					
2 5	-0 4021	-0 4021											

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TABLE A6. (Continued)

RUN SEQ	67 4	CLAERO = 1 0591	CDAERO = 0 5121	DCLC = 0.0	DWLC = 1.1202	BASEPR = 8.1904	PAGE 109				
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.00	2118.83	2111.71	7 12	76.68	0 01	2 66	0.0	2.003	2.003	87.0827	0.508079E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	2 18	-1.15	-1.25	0.00	-0.01	0.02	1.15	-2.47	1 14	0.21	

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10 125
%X/C	CP	CP
0 0	-0.4851	-0 4851
2 5	-0 4507	-0 4507
5 0	-0.4851	-0 4507
10 0	-0 4507	-0.4851
15 0	-0 4851	-0 4851
25 0	-0 4851	-0.4851
35 0	-0.4851	-0.4507
50 0	-0 4507	
56 0	-0.4851	-0 4507
65 0	-0.4507	-0.4851
76 0	-0 4851	-0.4507
79 0	-0 4507	-0 4507
80 5	-0 4465	-0.3967
81 0	-0.4465	-0 4216
82 0	-0 4465	-0.4216
84 0	-0 4465	
87 0	-0 4216	
89 0	-0 4378	-0 4378
93 0	-0.4378	
96 0	-0 4378	-0 4378
100 0	-0 4378	-0 4010
96 0	-0.4746	-0 4378
84 0	-0.4378	-0.4010
73 0	-0.4746	-0 4378
50 0	-0 4746	-0.4378
35 0	-0.4851	-0.4851
25 0	-0.4851	-0.4851
10 0	-0.4851	-0.4851
5 0	-0 4851	-0 4851
2.5	-0 4507	-0 4851

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0 2597	-0 9848	-2 0484	-2.8702	-3.1119
2 5	-0.8399	-1.1299	-1 5167	-1.6616	-2 9186
5 0	-0 7916	-0 9365	-1.1782	-1.4683	-1.6133
10 0	-0 6948	-0.8399	-1.0816	-1 2265	-1 2750
15 0	-0.6465	-0.7916	-0.9365	-1.0816	-1.1299
24 0	-0 5982	-0 6948	-0 8399	-0.8882	-0 9365
33 0	-0 6801	-0 6801	-0 9595	-0.9595	-0.9595
54 0	-0.8198	-0 9595	-0 9595	-3.0560	-1.0993
65 0	-1.1000	-1.2104	-1.2104	-1.3208	-1.4312
78 5	-2.7189	-2.9029	-3.0869	-3.3077	-3.6388
79 5	-19 7623	-47.5401	-91.0998	-73 7518	-80 8205
80 5	-18.5419	0 8727	7.7924	29.4982	-7.4409
81 3	16.5784	208 0168	7.3693	-8.6357	-12 8672
82 0	-8.4366	-5.6737	5.5772	-14 3860	-6.2711
84 0	2 6648	-0 6705	0.9225	-8 5361	-0 9692
87 0	4 5068	-3.9064	-2.7864	0 9474	-0.4713
89 0	-0 5302	-0 5601	-3.6745	-2.8960	1.3565
93 0	-0 0809	0 2485	-0.5601	-0.5900	-1 8778
96 0	-0.1109	-1.3088	1.2967	2.0453	-3 2254
100 0	-3 1355	-2.9558	-2.5067	-2.4467	-2 5665
96 0	0 0688	0 1287	-0.0210	-0.0809	-0.0809
84 0	0 3982	-0.7997	1 7159	0.3683	-0 5001
73 5	0 1586	0 2983	0.2983	0.2983	0.1586
54 0	0 0188	0.0188	0.0188	0 0188	-0 1209
33 0	0.0188	0.0188	0.0188	0.0188	-0.1209
24 0	-0 0180	-0.0180	0.0303	0 0787	-0.0663
10 0	0 1270	0.0787	0 2237	0.2237	0.1586
5 0	0 2237	0.2237	0 2720	0.3687	0.2983
2.5	0 3204	0.3204	0 4171	0 4171	0 2983

\*\*FUSELAGE\*\*

-----TOP----
XLOC CP
38.2 -0.1802
43.4 -0.1802
50.4 -0.1802
56.1 -0.2170
62.5 -0.4378
69.7 -0.4378
76.9 -0.3642
83.4 -0.3274
89.4 -0.4378
95.4 -0.4378
101.1 -0.4010
---BOTTOM---
38.2 -0 1066
43 4 -0.0698
50.4 -0.1802
56.1 -0.0330
62.5 0.0038
69.7 0.0405
76.9 0.0405
83 4 0.0774
89.4 0.0774
95.4 0.0774
101.1 -0 0698

\*\*MISC.\*\*

NO.	CP
1	-0.8198
2	0 0188
3	-0.9595
4	0 0188
5	-0.9595
6	-0.2607
7	0.3982
8	-0.7397
9	-0 3505
10	0.7875
11	0.0688
12	-0.2007

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TABLE A6. (Concluded)

RUN SEQ	67 6	CLAERO =	1 3178	CDAERO =	0 6275	DCLC =	0 0	DWLC =	1 2361	BASEPR =	8.1904	PAGE 110
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
8 01	2118 90	2111 78	7 12	76 66	0 01	2.66	0 0	2.007	2 007	87.1003 0	508532E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR	
-0 01	2 55	-0 95	-1 33	0 00	-0.01	0.02	1.45	-2.56		1.42	0.33	

***** CANARD *****	***** WING *****	**FUSELAGE**	**MISC.**
BP=3 375 BP=10 125	BP = 2 BP = 6 BP =12 BP =16 BP = 22	-----TOP----	NO. CP
%X/C CP CP	%X/C CP CP CP CP	XLOC CP	
0 0 -0 4507 -0 4161	0.0 -1 6131 -3 4499 -5.2383 -3 7882 -2.2898	38.2 -0.1434	1 -0 9594
2 5 -0 4161 -0 4507	2 5 -1 3232 -1 8065 -2 9183 -3.4016 -1.9999	43.4 -0 1434	2 0 1585
5 0 -0 4507 -0 4507	5.0 -1 0815 -1 4198 -2 0482 -3.3049 -1.9514	50.4 -0.1802	3 -1 7979
10 0 -0 4507 -0 4507	10 0 -0 9847 -1.2264 -1 5648 -3 2565 -1.9514	56 1 -0.2170	4 0 0188
15 0 -0 4851 -0 4507	15.0 -0 8881 -0 9847 -1 3232 -2.2415 -1 8548	62 5 -0.4010	5 -1 5183
25 0 -0 4507 -0 4507	24.0 -0 8398 -0 8881 -1 0815 -1.3232 -1 8065	69 7 -0 4010	6 -0 2607
35 0 -0 4507 -0 4507	33 0 -0 9594 -0 9594 -1 0991 -0.9594 -1 6582	76.9 -0 4010	7 0.3682
50 0 -0 4507	54.0 -0.9594 -0 9594 -1.0991 -2 9159 -1 6582	83.4 -0 3642	8 -0 7996
56 0 -0 4851 -0 4507	65.0 -1.1000 -1.2838 -1 3574 -1.3574 -1 6517	89.4 -0.4377	9 -0 4104
65 0 -0 4507 -0 4161	78.5 -2 7554 -2 9761 -3 1969 -3.1969 -3 8590	95.4 -0.4377	10 0 7575
76 0 -0 4507 -0 4161	79 5-19 6852 -47.4098 -91 0632 -73.7162 -80.5845	101.1 -0.3642	11 0.0388
79 0 -0.4507 -0.4507	80 5-19.5602 0.5988 7 4181 29.5179 -7.1662	---BOTTOM---	12 -0.2606
80 5 -0.4216 -0 4216	81.3 16.8007 204 6313 6 6714 -9.0825 -13 0893	38.2 -0.0699	
81 0 -0 4216 -0 3967	82.0 -8 7589 -5.7476 5.5515 -13.3631 -5.9217	43.4 -0.0331	
82 0 -0 4216 -0 3967	84 0 2.5899 -0 6456 0.9472 -8 7341 -1.1931	50 4 -0.1434	
84 0 -0 4216	87.0 4.4813 -3 9307 -2 8109 0.8975 -0 9691	56.1 0 0037	
87 0 -0 3967	89.0 -0 5601 -0 6199 -3 7639 -3.0153 0 8174	62 5 0.1141	
89 0 -0 3642 -0 4010	93 0 -0 1109 0 2484 -0.5900 -0 6199 -2 7759	69.7 0.1509	
93 0 -0 4010	96 0 -0 1109 -1 3685 1 1467 1 9552 -4 3029	76 9 0 1141	
96 0 -0 4010	100 0 -3.0454 -2 8957 -2.4165 -2.4464 -2 5363	83 4 0.1509	
100 0 -0 4377 -0 4010	96 0 0 1287 0 1885 0 0388 -0 0809 -0 1109	89 4 0 1509	
96 0 -0 4010 -0 4377	84 0 0 4281 -0 6199 1 6258 0 3682 -0.4104	95 4 0.1509	
84 0 -0 4377 -0 4010	73 5 0 1585 0.2983 0 2983 0 2983 0.0188	101.1 0 0037	
73 0 -0 4010 -0 4377	54 0 0.0188 0 0188 0 1585 0.0188 -0 2607		
50 0 -0 4010 -0 4010	33 0 0 1585 0 1585 0 1585 0 0188 -0 1210		
35 0 -0 4507 -0 4507	24 0 0 1270 0 1753 0 1753 0 1753 -0 0181		
25 0 -0 4507 -0 4161	10 0 0 2720 0 3203 0 3686 0 3686 0 1585		
10 0 -0 4507 -0.4507	5 0 0 4170 0 4170 0 4653 0 4653 0 2983		
5 0 -0 4507 -0 4507	2 5 0 4653 0 4653 0 5136 0 4653 0 2983		
2 5 -0 4507 -0 4507			

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TABLE A7. RUN 75, BC1W6V, DELF=45, CMU=0, (BN/B)=1

RUN SEQ		PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 133				
75 10	CLAERO = 0 5282	CDAERO = 0 2099	DCLC = 0 0	DWLC = 0 0	BASEPR = 8.1914					
ALPHA	PTOT	PSTAT	Q VEL	YAW H/C	CMUC	CMUW	CMUT	HGT	RN	
0 06	2133.33	2103 27	30.06	156.41	0 01 2 67	0.0	0 0	0 0	87.4624 0.106414E+07	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0 01	0.53	0 21	-0.18	0 00	-0 00	0 01	0 52	-0.18	0.52	0.21

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3 375	BP=10.125
%X/C	CP	CP
0 0	0.4671	0 5325
2 5	-0.0638	-0 2435
5 0	-0 1291	-0 1945
10 0	-0 0801	-0 2271
15 0	-0 2435	-0 2598
25 0	-0.2353	-0.2516
35 0	-0 2190	-0.2190
50 0	-0 2271	
56 0	-0.2353	-0 2598
65 0	-0 3251	-0 3170
76 0	-0 2108	-0 5130
79 0	-0.4885	-0 6600
80.5	-0 4524	-0.6352
81 0	-0.4406	-0.6293
82 0	-0.4524	-0 6234
84 0	-0.2107	
87 0	-0.4642	
89 0	-0.4662	-0.6666
93.0	-0 4836	
96 0	-0 4836	0 6320
100 0	0 9457	-0 6492
96 0	0.3444	0 3444
84 0	0.9457	-0 6492
73 0	0 3444	0.3444
50 0	0 6756	0.5710
35.0	0 5570	0 5488
25 0	0 1813	-0.3415
10 0	0 0506	0 0506
5.0	-0.0801	0 0016
2 5	-0 1046	-0 0393

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	0.0132	0.0705	0.3224	0.1736	-0.3074
2 5	0 1965	0 1965	-0 2616	-0 5364	-0.6967
5.0	0 1392	0.1049	-0 2043	-0 4448	-0.6166
10 0	0.0018	-0.0325	-0 2501	-0.4104	-0 4677
15 0	-0.0440	-0 0784	-0 2845	-0.3188	-0 4334
24 0	-0 1127	-0.1470	-0 2959	-0.2959	-0.3646
33.0	-0 1276	-0 1938	-0 3262	-0.2931	-0.2931
54 0	-0.2600	-0.2931	-0 3593	-0 3262	-0.3262
65 0	-0.3442	-0 3877	-0 4488	-0 4226	-0 4139
78 5	-0.6666	-0 6928	-0 7364	-0.7364	-0.7799
79 5	-0 5821	-0 5821	-0 6410	-0.6764	-0.7236
80 5	-0.5526	-0 5585	-0 6352	-0 6646	-0.7295
81 3	-0 5762	48.8672	-0 6470	-0 6823	-0.7354
82 0	-0 5644	-1.0125	-0.6410	-0.6705	-0 7295
84 0	-0.5644	-0 5939	-0.6352	-0 6764	-0.7413
87.0	-0 5939	-0 5998	-0.6470	-0.6764	-0.7236
89.0	-0 6477	-0.6477	-0.6832	-0.6193	-0.7115
93.0	-0 7044	-0.6548	-0 6832	-0.2505	-0.7896
96.0	-0 7044	-0.6335	-0.6832	-0.7115	-0.7967
100 0	-0 5626	-0.5697	-0.5626	-0.7115	-0.7612
96.0	0.2035	0 2887	0.2957	0.3028	0 2816
84.0	0 3880	0.5299	0.5795	0.5653	0.5228
73 5	0 3690	0.4021	0 5014	0 5345	0.4683
54 0	0 1704	0.2035	0 2366	0.2366	0.2035
33.0	0.0049	0 0049	0 0711	0.1042	0 0711
24.0	-0 0440	-0 0096	0 0362	0.0705	0.0362
10 0	-0 0784	-0.0898	0 0132	0.0820	0.1042
5 0	-0.1127	-0 1585	0 0132	0 1277	0 1704
2 5	-0 1814	-0 2845	-0 0440	0.2079	0.2035

\*\*FUSELAGE\*\*

----	TOP----	**MISC.***
XLOC	CP	NO. CP
38.2	-0 1524	1 -0.2600
43.4	-0.1524	2 0.0711
50.4	-0.1350	3 -0.3262
56.1	-0.0827	4 0.0711
62.5	-0.2047	5 -0 2931
69.7	-0.2047	6 0.0049
76.9	-0.0914	7 0.5157
83.4	-0.0827	8 -0.3143
89 4	-0.1698	9 -0 1228
95 4	-0.2221	10 0.6008
101.1	-0.1786	11 0.2745
----	BOTTOM----	12 -0.0518
38.2	0.0045	
43.4	0.0132	
50 4	-0.0391	
56.1	0.0219	
62.5	0.0132	
69.7	0.0132	
76 9	0 0480	
83 4	0 1090	
89 4	0.1439	
95.4	0.0567	
101.1	-0.3703	

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TABLE A7 (Continued)

RUN SEQ 75 12      CLAERD = 0 7283    CDAERD = 0 2632    DWLC = 0 0      BASEPR = 8 1914      PAGE 135  
 ALPHA PTOT      PSTAT    Q    VEL      YAW    H/C    CMUC    CMUW    CMUT      HGT      RN  
 4 01    2133 26    2103 08    30 17 156 90    0 01 2 67    0 0    0 0    0.0      87.4835 0.106275E+07  
 BETA    CL      CD      CM      CROLL    CN      CY      CNTR    CMTR      CLTR    CDTR  
 -0 01    0 73    0 26    -0.14    0 00    -0 00    0 00    0 74    -0.14    0 72    0 26

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
BP=3 375    BP=10 125			BP = 2		BP = 6	BP =12	BP =16	BP = 22	-----TOP-----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLCC	CP	
0 0	0 1546	-0 6102	0 0	0 1453	0 2480	-0 2996	-1.3833	-2.0678	38 2	-0 1886	1 -0 3513
2 5	-0 6021	-1 1798	2 5	0 0769	-0.0486	-0 7217	-0.9841	-1 4404	43.4	-0 1886	2 0 1434
5 0	-0 5370	-0 6916	5 0	0 0084	-0 1057	-0 5277	-0.7787	-1.0411	50.4	-0.1539	3 -0.4503
10 0	-0 0650	-0 5532	10.0	-0 0943	-0 1855	-0 4821	-0 6304	-0 7673	56.1	-0 0931	4 0 1764
15 0	-0 4637	-0 5289	15 0	-0.1513	-0 2311	-0 4251	-0 5620	-0 6418	62 5	-0.1973	5 -0 4173
25 0	-0 3742	-0 3986	24 0	-0 2083	-0 2768	-0 4023	-0.4365	-0 4593	69.7	-0.1973	6 0 0115
35 0	-0.3092	-0.3417	33 0	-0 2524	-0 3183	-0 3843	-0.4173	-0.4503	76.9	-0.1192	7 0 3931
50 0	-0 3010		54 0	-0 3183	-0 3513	-0 4173	-0.3183	-0.4173	83 4	-0.1105	8 -0 4408
56 0	-0 3092	-0 3336	65 0	-0 4231	-0.4491	-0 4838	-0 4578	-0.4752	89 4	-0 1886	9 -0.4761
65 0	-0 3905	-0 4312	78 5	-0.7269	-0.7356	-0.7617	-0.7617	-0.8311	95.4	-0.2060	10 0.4497
76 0	-0 2196	-0 5777	79 5	-0 6144	-0 6203	-0 6673	-0 7143	-0.7083	101.1	-0 1713	11 0.4497
79 0	-0 5614	-0.6997	80.5	-0.5909	-0 6026	-0 6614	-0.7201	-0 6790	---BOTTOM---		12 -0.1439
80 5	-0 5498	-0 6849	81 3	-0.6261	44 3310	-0 6614	-0 6790	-0.6849	38.2	0.0371	
81 0	-0 5380	-0 6966	82 0	-0.6144	-1 0256	-0 6731	-0.6966	-0.6966	43.4	0.0545	
82 0	-0 5321	-0 6849	84 0	-0.6261	-0 6379	-0 6790	-0 6849	-0 6614	50.4	-0.0063	
84.0	-0 2267		87 0	-0 6731	-0.6555	-0 6849	-0.6731	-0.5733	56.1	0.0458	
87 0	-0 5380		89 0	-0.7517	-0 6881	-0.7305	-0.7022	-0 7729	62 5	0.0545	
89 0	-0 5273	-0 7269	93 0	-0.7800	-0.6810	-0.7305	-0 2641	-0.8577	69 7	0.0632	
93 0	-0 5533		96.0	-0.7517	-0 6810	-0.7234	-0.7659	-0 8506	76.9	0.0892	
96 0	-0 5359	0 6709	100 0	-0.6245	-0 5892	-0.5962	-0 7517	-0 8295	83.4	0.1500	
100 0	0 8012	-0.7356	96 0	0 2447	0 3295	0 3578	0.2942	0 2518	89 4	0 1934	
96 0	0 3149	0.3236	84 0	0.4709	0.5981	0 6051	0 5698	0 4709	95.4	0.0892	
84 0	0 8012	-0 7356	73 5	0.4072	0.4732	0.5392	0.5392	0.4402	101.1	-0.3710	
73 0	0 3149	0.3236	54 0	0.2094	0 2094	0 2753	0.2753	0.2094			
50 0	0 7665	0 6102	33.0	0 0774	0 1104	0 1434	0.1764	0 0774			
35 0	0 6266	0 5859	24 0	0 0655	0.0769	0 1339	0 1795	0.1225			
25 0	0 2035	-0 4068	10 0	0.0198	0 0198	0 1795	0.2594	0 2423			
10.0	0.1384	0 1384	5 0	-0 0144	0 0198	0.2366	0.3392	0 3413			
5 0	0 0326	0 1221	2 5	-0 0144	-0.0600	0.3278	0.4305	0 4402			
2 5	0 2767	0 3581									

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TABLE A7 (Concluded)

RUN SEQ	CLAERO =	1.1935	CDAERO =	0.4592	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1914	PAGE	137
75 16	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	HGT	RN		
12 08	2133.54	2103.48	30 06	156 80	0.01	2 66	0 0	0 0	87.1285	0.105731E+07		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0 01	1.19	0 46	-0 06	0 00	-0 00	0 00	1.25	-0.06	1.18	0.46		

\*\*\*\*\* CANARD \*\*\*\*\*

%X/C	BP=3 375 CP	BP=10.125 CP
0.0	-5 0868	-1 7054
2 5	-2 3427	-1 5748
5 0	-1 6074	-1 5748
10 0	-0 0556	-1 5339
15 0	-1 0684	-1 5094
25 0	-0 7988	-1 4441
35 0	-0 6518	-1 3379
50 0	-0 5375	
56 0	-0 5130	-0.9050
65 0	-0.5538	-0 7335
76 0	-0 3170	-0 7417
79 0	-0 6763	-0.8805
80.5	-0 6764	-0 9181
81.0	-0 6469	-0 8887
82 0	-0 6410	-0 9004
84 0	-0 3168	
87 0	-0 6764	
89 0	-0 6753	-0 9281
93.0	-0 6928	
96 0	-0 6840	0 7278
100.0	0 6843	-0 8845
96.0	0 3356	0 2833
84 0	0 6843	-0 8845
73 0	0 3356	0 2833
50 0	0 7888	0 6494
35 0	0 6876	0 5815
25 0	0 3283	-0 5865
10 0	0 3038	0 2629
5 0	0 2629	0 3201
2 5	0 6060	0 5815

\*\*\*\*\* WING \*\*\*\*\*

%X/C	BP = 2 CP	BP = 6 CP	BP = 12 CP	BP = 16 CP	BP = 22 CP
0 0	0 0018	-1 0746	-4 4754	-3 6853	-1 5784
2 5	-0.4104	-0.9142	-2.0593	-3.5363	-1 5555
5 0	-0 3875	-0 7196	-1 3723	-3 5592	-1 5097
10 0	-0 4562	-0.6623	-1.0631	-1.7845	-1.5211
15.0	-0.4906	-0 6165	-0 8914	-0.9142	-1 4982
24 0	-0.4906	-0 5707	-0 7196	-0.7196	-1 4410
33 0	-0 4917	-0 5579	-0 6573	-0.6242	-1 4518
54 0	-0.4586	-0.5248	-0 5911	-0.3262	-1.1538
65.0	-0 5359	-0.5708	-0.5969	-0 5882	-0 8584
78.5	-0 7973	-0.7973	-0.8148	-0.8322	-0.9804
79.5	-0 7000	-0.6823	-0.7236	-0.7295	-0.7413
80 5	-0.6882	-0.6528	-0.7118	-0.7295	-0 6823
81.3	-0 7000	40.0000	-0.7118	-0.7118	-0.7413
82 0	-0.6882	-1 0184	-0.7118	-0.6351	-0.7825
84 0	-0 7118	-0.6587	-0.7177	-0.6351	-0.8061
87 0	-0 7236	-0 7295	-0.7059	-0 5880	-0.6764
89 0	-0 8179	-0 7896	-0.7966	-0 7612	-0.8818
93.0	-0 8534	-0.7825	-0 8108	-0.3569	-0.9669
96 0	-0.7966	-0.7470	-0 8038	-0 8250	-0.9669
100 0	-0.6548	-0 6264	-0 6264	-0 8179	-0 9385
96.0	0 2957	0.4234	0 3880	0.2745	0 2106
84 0	0 5937	0.7143	0.6433	0.5724	0 3809
73 5	0 5345	0 6007	0 6007	0.5676	0 3690
54.0	0 2697	0 3359	0.3359	0.3359	0 2035
33 0	0 2035	0 2697	0 3028	0.3028	0 2035
24.0	0 1965	0 2652	0 3224	0.3453	0 2423
10 0	0 2079	0.2995	0.4255	0 4484	0 4021
5 0	0 2308	0 3339	0.4484	0.5056	0 4352
2.5	0 2652	0.3568	0.4255	0 4598	0.4352

\*\*FUSELAGE\*\*

XLOC	CP	**MISC.**
----	----	----
38.2	-0 3180	1 -0 6242
43.4	-0.2831	2 0.3028
50.4	-0.2134	3 -1 1207
56 1	-0.1611	4 0.2697
62 5	-0.3006	5 -1 2200
69 7	-0.2919	6 0 0711
76.9	-0.2222	7 0 2816
83.4	-0.1960	8 -0.4845
89.4	-0.2483	9 0 4376
95 4	-0 2570	10 0 0120
101.1	-0.1960	11 -0 7470
----	----	----
38.2	0.1090	12 -0 3001
43.4	0.1178	
50.4	0.0306	
56.1	0.1178	
62.5	0.1526	
69.7	0.1962	
76 9	0.2136	
83.4	0.2398	
89.4	0.2659	
95.4	0.1352	
101 1	-0.3790	

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TABLE A8 RUN 76,BC1W6V, DELF=45, CMU=0 5, (BN/B)=1

RUN SEQ	76 2	CLAERD =	0 9376	CDAERO =	0.6596	DCLC =	0.1844	DWLC =	0 4560	BASEPR =	8.1914	PAGE	138
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
-0 01	2133.54	2106.65	26 89	148 62	0.01	2.66	0.225	0 527	0.752	87.0591	0.994008E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR		
-0.01	1 58	0 27	-0 51	0 00	-0 01	0 03	1.05	-0 30		1 05	0.52		

\*\*\*\*\* CANARD \*\*\*\*\*

%X/C	BP=3 375 CP	BP=10.125 CP
0 0	0 2425	-1 0811
2 5	-0 5517	-1.4006
5 0	-0 5152	-0 9442
10 0	-0 0679	-0 7708
15 0	-0 5334	-0 7982
25 0	-0 5243	-0 6612
35 0	-0.5334	-0 6612
50 0	-0.6795	
56 0	-0 8073	-0 9442
65 0	-1 1633	-1 4098
76 0	-0 3783	-3 0712
79 0	-4 4770	-4 8879
80 5	-22 1755	-19 2103
81 0	-19 4605	-18 4194
82 0	-14 4592	-16.1591
84 0	-0 3243	
87 0	0 0447	
89 0	-0 9382	-1 5032
93 0	-0 7824	
96 0	-0 3343	0 6787
100 0	1 2047	-0 0518
96 0	0 6495	0 5813
84 0	1 2047	-0 0518
73 0	0 6495	0 5813
50 0	0 7859	0 5424
35 0	0 6898	0 5802
25 0	0 3612	-1 0994
10 0	0 2425	0 2060
5 0	0 0964	0 1877
2 5	0 1786	0 4342

\*\*\*\*\* WING \*\*\*\*\*

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0 0	-0 4588	-0 4460	-0 8811	-0 9835	-3 0697
2 5	0 6802	0 6162	-0 8427	-0 9579	-1 8410
5 0	0 5266	0 4626	-1 0219	-0 8427	-1 4058
10 0	0 2963	0 1811	-0 9707	-0 8044	-1 0603
15.0	0 1427	0 0147	-1 0347	-0 7916	-0.9067
24 0	-0 0749	-0 2028	-0 7916	-0.7787	-0.8044
33 0	-0 2209	-0 3319	-0 8869	-0 8869	-0 8128
54 0	-0 6279	-0 7018	-0 8869	0.2602	-1.1089
65 0	-1 0162	-1 1136	-1 1428	-1.5812	-1.5812
78 5	-3 1593	-3.2370	-2 9351	-3 9870	-5.0002
79 5	-21.9119	-22.1030	-13 5233	-29.1210	-37 2448
80 5	-22 2614	-20 1788	-14 3670	-21 2991	-17 8595
81 3	-21.7538	44 1829	-14 9663	-14.2350	-6 8414
82 0	-19 0783	-17.1675	-14 6503	-11.3620	-3.5862
84 0	0 0183	-5 3062	-13.7342	-10 1759	-3 1645
87 0	0 4071	-1 0426	-2.7164	-5 2731	-2.8154
89 0	-0 7837	-0 6172	0.6910	-3 3047	-2 2108
93 0	-0 6330	-0 6172	-0 2604	-0 5300	-3.1779
96 0	-0 6330	-0 7123	-0 3873	-0.7123	-3 0987
100 0	-0 3555	-0 2287	-0 1336	-0 0701	-0 3635
96 0	0 4135	0 2391	0 3342	0 4214	0 4373
84 0	0 3580	0 2708	0 2312	0.4848	0 5086
73.5	0 1492	0 1492	0 2602	0 3712	0.3712
54.0	-0 1468	-0 0728	-0 0358	0 1492	0.0752
33 0	-0 2949	-0.3319	-0.2949	-0.0728	-0 0358
24 0	-0.3308	-0 3180	-0.3436	-0 1388	0 0019
10 0	-0 4716	-0 3948	-0.3948	-0 1900	0.1492
5 0	-0 4588	-0.3308	-0 4588	-0.2540	0 2602
2 5	-0 4332	-0 3692	-0 5228	-0.0749	0.3712

\*\*FUSELAGE\*\*

XLOC	CP
38.2	-0.3343
43 4	-0 3538
50.4	-0 2174
56 1	-0.1103
62 5	-0.3733
69 7	-0 3733
76.9	-0 1687
83 4	-0 2272
89 4	-0 3733
95.4	-0 3538
101.1	-0 2759
---BOTTOM---	
38.2	0.1527
43.4	0.2014
50.4	0.1138
56 1	0.1138
62.5	-0.0616
69.7	-0.1980
76 9	-0.1590
83.4	-0.0129
89 4	0.1040
95 4	0 0456
101 1	-0 2662

\*\*MISC \*\*\*

NO.	CP
1	-0 4428
2	-0.3319
3	-0 8128
4	0 0011
5	-0.8499
6	-0.1468
7	0 3025
8	-0.8233
9	-0 5220
10	0 8495
11	0 1122
12	-0 2366

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TABLE A8 (Continued)

RUN SEQ 76 4      CLAERO = 1 1080    CDAERO = 0 7649    DCLC = 0 1868    DWLC = 0.4619      BASEPR = 8.1914      PAGE 140  
 ALPHA PTOT      PSTAT      0    VEL      YAW    H/C      CMUC      CMUW      CMUT      HGT      RN  
 3 99    2133 61    2106.04 27 57 150 58    0 01 2.65    0 218    0 514    0 732    86.7386 0.100513E+07  
 BETA    CL      CD      CM      CROLL    CN      CY      CNTR    CMTR    CLTR    CDTR  
 -0.01    1 76    0 43    -0.49    -0 02    -0 01    0.06    1.26    -0.29    1.22    0.64

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC ***	
%X/C	BP=3.375	BP=10.125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP
	CP	CP		CP	CP	CP	CP	CP	XLOC		
0 0	-1.0737	-4.7686	0 0	-0.4096	-0.4596	-1.2959	-1.6205	-5.7648	38.2	-0.3957	1 -0.5870
2 5	-1.3853	-2.2757	2.5	0.6265	0.5141	-1.0837	-1.3458	-2.8436	43.4	-0.3957	2 -0.1539
5 0	-0.8867	-1.7503	5 0	0.4766	0.3393	-1.1836	-1.0962	-2.0199	50.4	-0.2627	3 -0.8757
10 0	-0.0675	-1.2695	10 0	0.2145	0.1147	-1.4082	-0.9589	-1.3334	56.1	-0.1392	4 0.0626
15 0	-0.7976	-1.2072	15 0	0.0772	-0.0726	-1.3583	-0.8840	-1.0837	62.5	-0.4147	5 -0.9479
25 0	-0.7175	-0.9490	24.0	-0.1475	-0.2598	-1.2709	-0.8965	-0.9339	69.7	-0.4052	6 -0.1178
35 0	-0.6730	-0.8598	33 0	-0.2983	-0.4065	-1.2727	-0.9840	-0.9118	76.9	-0.2057	7 0.1813
50.0	-0.7887		54 0	-0.6592	-0.6953	-1.1283	0.6400	-1.1644	83.4	-0.2437	8 -1.0560
56.0	-0.9134	-1.1182	65.0	-1.0418	-1.0988	-1.3078	-1.5643	-1.4788	89.4	-0.3672	9 -0.7313
65 0	-1.2784	-1.6079	78.5	-3.1414	-3.2270	-2.9513	-3.3412	-4.7661	95.4	-0.3767	10 0.8772
76 0	-0.4059	-3.2552	79 5-20	7942	-21.0387	-12.9401	-30.3645	-36.2189	101.1	-0.3007	11 -0.0121
79.0	-4.5372	-4.9826	80.5-21	5913	-19.0846	-12.9143	-19.6950	-16.6229	---	---	12 -0.3755
80 5	-21.0771	-15.9415	81.3-21	3277	35.4838	-14.0070	-10.9991	-5.6771	38.2	0.1649	
81 0	-19.5539	-16.4042	82 0-19	2003	-17.2594	-14.4310	-8.4217	-3.1638	43.4	0.2218	
82 0	-15.8707	-16.0252	84.0	-0.6702	-6.3520	-14.0389	-1.6407	-2.7849	50.4	0.1174	
84 0	-0.5673		87.0	0.4353	-0.9594	-2.9133	-1.2551	-2.4313	56.1	0.1459	
87 0	0.2554		89.0	-0.6848	-0.6462	0.4828	-1.7907	-2.0072	62.5	0.0033	
89 0	-0.6522	-2.2294	93.0	-0.6385	-0.6694	-0.2982	-0.5457	-2.9582	69.7	-0.0917	
93 0	-0.7473		96 0	-0.6771	-0.6694	-0.5920	-1.1179	-2.8345	76.9	-0.0252	
96 0	-0.3577	0.7824	100 0	-0.2441	-0.0121	-0.1281	-0.3369	-0.3369	83.4	0.1174	
100 0	1.5520	-0.0632	96 0	0.6143	0.3050	0.2663	0.2586	0.3900	89.4	0.2313	
96 0	0.6779	0.5449	84 0	0.3978	0.3746	0.2895	0.4442	0.4442	95.4	0.1459	
84 0	1.5520	-0.0632	73.5	0.2070	0.2791	0.2431	0.3153	0.2791	101.1	-0.2532	
73 0	0.6779	0.5449	54.0	-0.0096	0.0265	0.0987	0.1709	0.0626			
50 0	0.7729	0.5734	33.0	-0.2261	-0.2261	-0.0457	0.0265	0.0265			
35 0	0.7338	0.5735	24 0	-0.2723	-0.3347	-0.1350	0.0148	0.1147			
25 0	0.4133	-1.2339	10 0	-0.3847	-0.3223	-0.3472	0.0397	0.2791			
10 0	0.3064	0.2708	5.0	-0.6093	-0.4471	-0.4346	0.0273	0.3874			
5 0	0.2263	0.2619	2 5	-0.5095	-0.3971	-0.5095	0.1022	0.3513			
2 5	0.4311	0.5469									

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TABLE A8 (Concluded)

RUN SEQ				PROPLULSIVE		WING / CANARD	PRESSURES			PAGE 142
76 10	CLAERO =	1.6791	CDAERO =	1 1623	DCLC =	0 2026	DWLC =	0.4902	BASEPR =	8.1914
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT RN
12 03	2133 54	2106 42	27.12	149 54	0 01	2.65	0 220	0 515	0 735	86 7934 0.993237E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0 01	2 37	0.92	-0 47	-0 02	-0 02	0.06	1 99	-0 26	1.81	1.05

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
%X/C	BP=3 375	CP	BP=10 125	CP	BP = 2	BP = 6	BP =12	BP =16	BP = 22	----	TOP----	NO	CP
0 0	-9.7371		-3 1739		0.0	-1.5264	-1.0568	-0.8283	-6.2730	-4.4202	38 2	-0.5753	1 -1.2757
2 5	-3.7624		-3 1831		2.5	0 4281	0 0220	-1.1583	-3.0622	-4.6230	43 4	-0.5463	2 0.3021
5 0	-2 3321		-3 2285		5.0	0 2251	-0 1557	-1.1076	-2.1483	-4 6359	50.4	-0 3724	3 -1.2757
10 0	-0 0690		-3.3642		10 0	-0 0288	-0 4222	-1.2598	-1 6913	-5 3973	56 1	-0.2566	4 0.3388
15 0	-1.5626		-3.3822		15 0	-0.2064	-0 4857	-1 3106	-1.4756	-4 6864	62.5	-0.4884	5 -2.9634
25 0	-1.2277		-3.9527		24 0	-0 3841	-0 7268	-1.4122	-1.3360	-2.3005	69 7	-0.5173	6 -0 1015
35 0	-1.1372		-3 3280		33 0	-0 5419	-0 7987	-1.5325	-1 3858	-1 2757	76.9	-0.3049	7 -0.1172
50.0	-1 1643				54 0	-0.8354	-1.0188	-1.7894	-0.0282	-1 6426	83.4	-0.3145	8 -1.7919
56 0	-1 2368		-1 0919		65 0	-1.1839	-1 3578	-2 0532	-1.9566	-2 1981	89.4	-0.4304	9 -1.4380
65 0	-1 6169		-1 1734		78 5	-3 4442	-3.5795	-3 9080	-4.0238	-6 5451	95.4	-0.4207	10 0 8813
76 0	-0 5126		-2 8480		79.5-23	0747	-21 9571	-18 8857	-11 7566	-34.7325	101.1	-0.3145	11 -0 3373
79.0	-4.9843		-4 5500		80.5-23	2052	-20.2647	-18.5004	-8 9727	-26 6166	----	BOTTOM---	12 -0.8484
80 5	-21.4542		-9 6917		81.3-22	8329	13.6241	-18.0299	-11.7566	-18.6309	38.2	0.2458	
81.0	-20.3757		-10 3318		82 0-20	5390	-18.6571	-16.9319	-11.5866	-14.4423	43.4	0.2844	
82 0	-16.6314		-8 5675		84.0	-0.8305	-7 1756	-14 0894	-11.0575	-6.3132	50 4	0.1492	
84 0	-0 7521				87 0	-0 0529	-1 2095	-2.5753	-8.9661	-3.0522	56.1	0.1974	
87.0	-0 1313				89 0	-1.1943	-1 2808	-1 3594	-5.1646	-2.3422	62 5	0.1299	
89 0	-0 8072		-6 6513		93.0	-1 0921	-0 9663	-1 1078	-0 7147	-3 2620	69.7	0.1685	
93.0	-0 9327				96.0	-1 0764	-0 9663	-1.2887	-0 5496	-2 9398	76.9	0.2651	
96 0	-0 5077		0 8350		100 0	0.2681	0 2445	0 1029	0 0479	-0 2430	83.4	0.3617	
100 0	1 4725		-0 6719		96.0	0 8341	0 7319	0.6690	0 6297	0 2366	89.4	0 3907	
96 0	0 7094		0 4486		84 0	0 5432	0 6612	0 6061	0 5589	0 0951	95.4	0 1974	
84 0	1 4725		-0 6719		73.5	0 4855	0 5956	0 5222	0 4488	0 1553	101.1	-0 3532	
73 0	0 7094		0 4486		54 0	0 3755	0 4122	0 4488	0 4488	0 0819			
50 0	0 8543		0 6032		33.0	0 1920	0 1920	0 3021	0.3388	0 1186			
35 0	0 8091		0 5738		24 0	0 0601	0 1362	0.3139	0 4027	0 2631			
25 0	0 5013		-1 5807		10 0	-0 2445	-0 1430	0 1870	0 3900	0 3388			
10 0	0 4651		0 3655		5 0	-0.5745	-0 9299	0 0474	0.3393	0 2654			
5 0	0 4289		0 4199		2 5	-0 9933	-1 2726	0 0220	0 1489	-0 0282			
2 5	0 5919		0 3837										

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TABLE A9. RUN 77, BC1W6V, DELF=48, CMU=1.0, (BN/B)=1

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 143				
77 2	CLAERO =	1.0790	CDAERO =	0.8478	DCLC =	0.3621	DWLC =	0.8896	BASEPR =	8.1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.10	2133.96	2120.29	13.67	105.96	0.0	2.78	0.442	1.026	1.468	90.9884	0.706432E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0 0	2 33	0.08	-0 78	0 00	0 00	0 01	1.27	-0 36	1.27	0.58	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **	
	BP=3.375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	----	TOP----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO	CP
0 0	0.1939	-1 4220	0.0	-0.5588	-0.4833	-0 6847	-0.6091	-3.2019	38.2	-0.4351	1	-0.5264
2.5	-0.6320	-1.6734	2.5	0.7502	0 7502	-0.7601	-1.2133	-1.9433	43.4	-0.4351	2	-0.5992
5 0	-0.5961	-1 0809	5.0	0 5488	0 5488	-0.7854	-1.0622	-1.6160	50.4	-0.2626	3	-0.9631
10.0	-0.0754	-0.9013	10.0	0.3474	0 1712	-0.9364	-0.9364	-1.2636	56.1	-0.1285	4	-0.1625
15 0	-0 6320	-0 8834	15.0	0.1460	-0 0301	-0.7854	-0.8860	-1.0874	62.5	-0.4159	5	-1.0358
25.0	-0.5961	-0.8295	24 0	-0.0805	-0.2567	-0.9364	-0.8860	-1.0119	69.7	-0.4351	6	-0.2353
35.0	-0.6499	-0.8295	33 0	-0 3080	-0 4536	-0 9631	-1.0358	-1.0358	76 9	-0.2051	7	0.2534
50 0	-0 8295		54.0	-0.7447	-0.8903	-1.1814	-1 1814	-1.3269	83.4	-0.2626	8	-0.9318
56.0	-0.9552	-1.1347	65.0	-1.2397	-1.3739	-1.4697	-1.8911	-1.8528	89.4	-0.3967	9	-0.6043
65.0	-1.4220	-1.7272	78.5	-4.0369	-4 3435	-4.2094	-5.2440	-6.1636	95.4	-0.3967	10	0.8304
76 0	-0 4524	-3 9177	79.5-45	0095	-54.0822	-36.4290	-61.8978	-67.8470	101.1	-0 3201	11	0.0351
79 0	-5.7132	-6.5394	80.5-44	7.111	-47.0440	-35.8074	-39.2416	-10 4300	----	BOTTOM----	12	-0 2456
80 5	-37.1033	-34.3557	81.3-36	7793	115 4234	-33.8240	-15.3291	-9.3023	38.2	0.1397		
81 0	-18.9447	-25.5295	82.0-21	0.0706	-0.5664	-30.1821	-7.1764	-4.7398	43.4	0.2164		
82 0	-5.6341	-15 4585	84 0	6.3289	-0 1776	-4.4288	-1 3959	-3 7807	50.4	0.0631		
84 0	-0 4238		87.0	0.1335	-3 5603	0 4964	-1.5125	-3 2881	56.1	-0.0135		
87.0	-0.9422		89.0	-1.5243	-2.0077	-1.8206	-2 8966	-2 3040	62.5	-0.3010		
89 0	-1 6803	-2 6575	93.0	-1 0253	-1.1033	-1.5555	-0.6043	-4.0349	69.7	-0.4351		
93.0	-0.9906		96.0	-1.0253	-1.2280	-0.9006	-0.9318	-4.3780	76.9	-0.3776		
96.0	-0.1668	0.6953	100 0	-0.6823	-0.6667	-0.5575	-0 6199	-0.7758	83.4	-0.2626		
100 0	1 7299	-0.5309	96.0	0 0351	0 0662	0.1286	0.4717	0 4717	89.4	-0.1093		
96 0	0 5995	0.5421	84 0	0.0819	0.0819	0 0662	0.6432	0.5497	95.4	-0.1093		
84 0	1 7299	-0 5309	73 5	-0 1625	-0 1625	0 1625	0 3469	0.3469	101.1	-0 4925		
73 0	0 5995	0.5421	54 0	-0.4536	-0 4536	-0 3808	-0.1625	-0 0897				
50 0	0 7528	0 5804	33 0	-0 5264	-0 5992	-0.5992	-0.4536	-0.1625				
35 0	0.7146	0 5889	24 0	-0.5084	-0.4833	-0 5336	-0.4077	-0 1308				
25 0	0 3555	-1 2784	10.0	-0 4077	-0 4833	-0.4581	-0 3071	-0 0170				
10 0	0 2298	0 1939	5 0	-0 4330	-0.4330	-0.5084	-0.2567	0.2014				
5 0	0 1221	0 2118	2 5	-0 4330	-0 4581	-0 5588	-0.2316	0 3469				
2 5	0.1939	0 4273										

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TABLE A9. (Continued)

RUN SEQ 77 4 CLAERO = 1 2778 CDAERO = 0 9521 DCLC = 0 3803 DWLC = 0 9274 BASEPR = 8.1914  
 ALPHA PTOT PSTAT Q VEL YAW H/C CMUC CMUW CMUT HGT RN  
 4 02 2133 82 2120 15 13 67 105.96 0 01 2 68 0.444 1.032 1 475 B7.5090 0.706407E+06  
 BETA CL CD CM CROLL CN CY CNTR CMTR CLTR CDTR  
 -0 01 2 59 0 27 -0 75 -0 00 -0 00 0 02 1 54 -0 33 1 50 0.70

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3.375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	NO.	CP	
0 0	-1 7811	-4 9231	0.0	-0.5840	-0 4581	-1.4398	-1.6412	-5 4422	38.2	-0 4925	1 -0.5264	
2 5	-1.6195	-3 2713	2.5	0.7250	0 7250	-1.3391	-1 4398	-2.9753	43 4	-0 4925	2 -0 5264	
5 0	-1.0270	-2 7506	5 0	0 5740	0 5236	-1.4901	-1.2636	-2.1950	50 4	-0 3010	3 -1.1086	
10 0	-0.0574	-1 4938	10 0	0 3222	0 1712	-1 6160	-1.1629	-1.5908	56 1	-0.1668	4 -0.1625	
15 0	-0 9193	-1.2963	15 0	0.1460	-0 0301	-1.6412	-1.1126	-1.2636	62 5	-0.4542	5 -1.1086	
25 0	-0 8295	-1.1168	24 0	-0 1057	-0 2819	-1.4901	-1.1629	-1.1126	69 7	-0.4734	6 -0.2353	
35 0	-0 7936	-1 0270	33 0	-0 2353	-0 4536	-1.3269	-1.1814	-1.0358	76.9	-0.2435	7 0.1754	
50 0	-0 9731		54.0	-0 7447	-0 8175	-1.2541	-0.8903	-1 3997	83.4	-0.2818	8 -1.0565	
56 0	-1 1168	-1 3143	65 0	-1 2397	-1.3547	-1.4505	-2.1785	-1.9294	89.4	-0.3776	9 -0.6978	
65 0	-1 5656	-1 9427	78.5	-4 0561	-4 4010	-4.0753	-5.2823	-6 2786	95.4	-0.4159	10 0.8927	
76.0	-0.4884	-4 2229	79 5-45	5536	-54.3411	-38 0883	-61.6645	-68.0152	101.1	-0.3393	11 0 0195	
79 0	-5 9107	-7 0236	80.5-45	2296	-47 4594	-35.0293	-33.3055	-10 7796	---	---BOTTOM---	12 -0.3236	
80 5	-37 5962	-35 8330	81.3-36	8444	115.4746	-34.3169	-9.4186	-9 5612	38.2	0.1781		
81 0	-20 0468	-27 3826	82 0-20	8.118	-0 9941	-31.1540	-0.6960	-4 8565	43.4	0.2356		
82 0	-6 3603	-17 4155	84.0	6.0437	0 0428	-5.2971	-2 1088	-3 9492	50.4	0.0823		
84 0	-0 4498		87 0	0 0428	-3.6511	0 6131	-1.5773	-3 2881	56 1	0.0631		
87 0	-1 0330		89 0	-1 5867	-2 1169	-1.9921	-3 2397	-2 5379	62.5	-0.2435		
89 0	-1 7953	-2 8874	93.0	-1 0409	-1 1812	-1 7894	-0.6199	-4.3312	69.7	-0.3776		
93 0	-1.0481		96 0	-1 0097	-1.3372	-1.1033	-1.2436	-4 7210	76 9	-0.3393		
96 0	-0 2435	0 7337	100 0	-0 7602	-0 4171	-0 1833	-0.4640	-0 7758	83 4	-0.1477		
100 0	1 6725	-0 5692	96.0	0 1442	0 1754	0.2846	0.4717	0 4249	89.4	-0.0135		
96 0	0 6379	0 4846	84 0	0 1910	0 1442	0.2378	0 4249	0 6276	95 4	-0.1285		
84 0	1 6725	-0 5692	73.5	-0 0170	-0 0170	-0.0170	0.2013	0 3469	101.1	-0.4734		
73 0	0 6379	0 4846	54 0	-0 3080	-0 3080	-0.2353	-0 1625	-0 0170				
50 0	0 7528	0 5995	33 0	-0 4536	-0.4536	-0 4536	-0 3080	-0.0897				
35 0	0 7326	0 5889	24 0	-0 4077	-0 4833	-0 4581	-0.3071	-0 0554				
25 0	0 3914	-1 4220	10 0	-0 3826	-0 4581	-0 6594	-0 3574	0 2013				
10 0	0 3016	0.2478	5 0	-0 3826	-0 3323	-0 7098	-0 3071	0 3469				
5 0	0 2118	0.2657	2 5	-0 4077	-0 3826	-0.7601	-0.2567	0 4197				
2 5	0 4632	0 5530										

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TABLE A9. (Concluded)

RUN. SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 147				
77 8	CLAERO =	1.9938	CDAERO =	1.4224	DCLC =	0 4308	DWLC =	1.0351	BASEPR =	8.1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11 98	2133 82	2120.94	12.88	102.86	0 01	2.62	0 468	1.088	1.557	85.5867	0.685633E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	3.46	0 90	-0.79	0 01	-0.01	0.00	2.45	-0.34	2.26	1.19	

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10.125
%X/C	CP	CP
0 0	-10.3360	-3 6480
2.5	-4.2387	-3 8196
5.0	-2.3524	-3.8577
10 0	-0 0849	-3.8767
15.0	-1.7426	-4.3149
25.0	-1.4187	-4.7723
35.0	-1.3044	-3.7052
50.0	-1.3997	
56.0	-1.5140	-1.0758
65 0	-1.9713	-1.7236
76 0	-0 6375	-4 6389
79 0	-6 6777	-7.6306
80.5	-39.5196	-40.0698
81 0	-21.0749	-31.0749
82 0	-6 2612	-19.6445
84 0	-0.6632	
87 0	-1.4334	
89.0	-2.1707	-3.7159
93.0	-1.3981	
96 0	-0.4831	0.7978
100.0	1.7941	-0.6864
96 0	0.6555	0 4319
84 0	1.7941	-0.6864
73.0	0.6555	0.4319
50.0	0 8385	0.5742
35.0	0.8106	0.5248
25 0	0.5058	-1 8951
10 0	0.4676	0.3343
5.0	0.4105	0.4295
2 5	0.5629	0.3343

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0 4681	-0 3078	-2 3648	-4 2347	-5.7841
2.5	0.4936	0.0127	-1 6168	-2.7655	-5.3567
5.0	0 3333	-0 1208	-1.5901	-2.3648	-5.0896
10 0	0 0662	-0.3612	-1.6168	-1.9908	-4.2080
15 0	-0.1208	-0.5215	-1 8037	-1.7770	-2.8456
24 0	-0 3880	-0.7352	-1.8572	-1.6702	-2.1777
33 0	-0.5435	-0 7752	-1.9336	-1.7020	-1.9336
54.0	-0.9296	-1 0841	-2.0881	-0.6207	-2.3198
65.0	-1.3777	-1.5404	-2.4147	-2.7807	-2.9026
78.5	-4.3259	-4 8342	-5.2205	-6.3998	-8.5751
79.5	-50.0970	-57.4971	-42.4493	-62 6550	-79 0764
80 5	-49.2309	-50.6612	-40.3865	-42.9996	-15.8347
81.3	-36 8653	106.9150	-38.2545	-16.9212	-12.1482
82.0	-16.3298	-2.9876	-34 2243	-7.3613	-6 5226
84 0	4.5912	0 4372	-0 4568	-1.7498	-5 5048
87 0	-0.3880	-4 3631	-1 5847	-2 4237	-4.1981
89.0	-1 9776	-2 6892	-3.6656	-3.7980	-3 1194
93.0	-1.3488	-1.7624	-2 5237	-0.8357	-4.8240
96.0	-1 3322	-1 9610	-1.5804	-1.6632	-5.0557
100.0	-0.5875	-0.4386	0.0082	-0.1076	-0 5213
96 0	0 6536	0.4054	0 3392	0.4385	0.4054
84 0	0 4882	0 3889	0.3061	0.3558	0.2730
73 5	0.2288	0.3061	0.3061	0 3833	0 2288
54 0	-0.0801	-0 0801	0.1516	0 1516	-0.0029
33.0	-0.3118	-0.3118	-0.1573	-0 0029	-0.0029
24.0	-0 3345	-0.3612	-0 2276	0.0127	0.1196
10 0	-0 3612	-0 3612	-0.4681	-0.1475	0 3061
5.0	-0 3612	-0 3612	-0.6016	-0.1475	0.3061
2 5	-0 3612	-0.3078	-0.7619	-0.2544	-0 0029

\*\*\*FUSELAGE\*\*

-----TOP----		**MISC.***
XLOC	CP	NO. CP
38.2	-0 7270	1 -1.3930
43 4	-0.6864	2 -0 2345
50.4	-0.4424	3 -1.7020
56.1	-0.2798	4 0.1516
62.5	-0.6458	5 -2.6287
69.7	-0.6255	6 -0.2345
76.9	-0.3408	7 -0.2896
83 4	-0.3815	8 -1.9942
89.4	-0.5035	9 -1.5970
95.4	-0.4831	10 0.9019
101.1	-0.4018	11 -0 5875
---BOTTOM---		12 -0 9185
38.2	0.2489	
43.4	0.3099	
50 4	0 1269	
56 1	0.1066	
62.5	-0.1171	
69.7	-0.1984	
76.9	-0.0561	
83.4	0.1269	
89.4	0.2285	
95.4	0.1269	
101.1	-0.5238	

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TABLE A10. RUN 79, BCIW6V, CLF=45, CMU=2.0, (BN/B)=1

RUN SEQ	PROPLULSIVE WING / CANARD PRESSURES										PAGE 148	
79 3	CLAERO =	0 8181	CDAERO =	1.1952	DCLC =	0 7695	DWLC =	1.8684	BASEPR =	8.1914		
ALPHA	PTOT	PSTAT	0	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
0 04	2131 63	2124.85	6 78	74.75	0 01	2 67	0 939	2.157	3 096	87.2223	0.494944E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0 01	3 46	-0 42	-1 27	-0 01	0 01	0 03	1 30	-0 41	1.30	0.60		

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
%X/C	BP=3 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP
	CP	CP		CP	CP	CP	CP	CP	XLOC	CP	
0 0	0 1339	-1 8936	0 0	-0.7738	-0 7230	-0 7738	-0 5707	-2.7026	38.2	-0.4653	1 -0.6115
2 5	-0 7712	-2 0747	2.5 0	0 7997	0 9012	-0 5200	-1.1799	-2.3981	43.4	-0.5039	2 -0.6115
5 0	-0.7712	-1 4229	5 0	0 6982	0.6475	-0 5200	-0.9768	-1 7382	50.4	-0.2721	3 -1 1984
10 0	-0 1196	-1 2057	10 0	0.4444	0 2414	-0 5200	-1.0275	-1 3320	56.1	-0.1176	4 -0.3180
15 0	-0 7351	-1.1333	15 0	0 2414	-0 0124	-0 5707	-1.0275	-1 2306	62.5	-0.5039	5 -1 1984
25 0	-0 7712	-1 0247	24.0	-0 0632	-0 3170	-0.7230	-1.0275	-1.0783	69.7	-0.5039	6 -0.3180
35 0	-0 8075	-0 9885	33.0	-0.3180	-0.4647	-0 9050	-1.1984	-1 1984	76 9	-0.1949	7 0 2481
50 0	-1.0247		54.0	-0 9050	-1 0517	-1 1984	-2.5191	-1.4919	83.4	-0 2335	8 -0.9467
56.0	-1.1696	-1 3868	65 0	-1.4312	-1.6243	-1.6630	-2.1652	-2 1652	89.4	-0.4267	9 -0.6009
65 0	-1.7125	-2 0747	78.5	-5.1012	-5.5262	-5.5262	-6 5306	-7 3420	95.4	-0.4653	10 0.9084
76.0	-0 5540	-5.0072	79 5-95	8533-112	3708	-90.5991-116.3685-	105 7585	101.1	-0.3494	11 0 0594	
79.0	-7.2519	-8 5553	80.5-89	8161	-84 8244	-78.1591	-63.8122	-23.4852	---	BOTTOM---	12 -0 2235
80 5	-44 8645	-55 7621	81 3-47	8949	154.8818	-48.7325	-15 0959	-12.5090	38 2	0 1528	
81.0	-12 8487	-35.7690	82.0	5.1843	26.0142	-26 6217	0.7414	-5.1651	43.4	0.2301	
82.0	-2.8914	-19 4609	84 0	3 6163	-3 2572	9 6272	-1 8460	-3 7539	50.4	0.0369	
84 0	-0 6437		87 0	0.9505	-5 5310	-3 1788	-1 7153	-3 3879	56.1	-0.1949	
87 0	-0 8789		89 0	-2.0788	-2.3618	-4 1855	-4 3741	-1.4813	62.5	-0.5812	
89 0	-2 3196	-3 3628	93 0	-1 1983	-0 9783	-2 0473	-0 7267	-4 5943	69.7	-0.6198	
93 0	-1 0061		96 0	-1 3241	-1 5757	-0 7581	-0 6009	-5.6320	76 9	-0 5425	
96 0	0 4619	0.7323	100 0	-1.1039	-1 1983	-1 3555	-1.4499	-1.5127	83.4	-0.4267	
100.0	1 3118	-1 4312	96 0	-0.1921	-0 0978	-0 0663	0 3110	0.4996	89 4	-0.2335	
96 0	0 5392	0 4619	84 0	-0 0978	-0 2550	-0.0978	0 7198	0.8141	95.4	-0.2721	
84 0	1.3118	-1 4312	73 5	-0 3180	-0 4647	-0 1712	0.4158	0.5625	101 1	-0.7357	
73 0	0 5392	0 4619	54 0	-0 6115	-0 6115	-0 6115	-0 3180	-0.0245			
50.0	0 7323	0 6551	33 0	-0.6115	-0 6115	-0 6115	-0 4647	-0 1712			
35 0	0.6769	0 5321	24 0	-0.5707	-0 6216	-0 5707	-0.6216	-0.3170			
25.0	0.2787	-1 5677	10 0	-0 5200	-0 5707	-0 5707	-0 5200	-0.0245			
10.0	0.2062	0 1339	5 0	-0.5200	-0 6216	-0 7230	-0.4692	0 2690			
5.0	0 0977	0 0977	2 5	-0.5707	-0 6723	-0 6216	-0 3677	0 2690			
2.5	0 1339	0 4235									

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TABLE A10. (Concluded)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 150				
79 11	CLAERO =	1 7438	CDAERO =	1 6927	DCLC =	0.8416	DWLC =	2.0290	BASEPR =	8.1914	
ALPHA	PTOT	PSTAT	O	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12 00	2131 56	2124 89	6 67	74 14	0 01	2 67	0 914	2.133	3.048	87.3492	0.490514E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	4.61	0 68	-1 26	0 01	0 00	-0 01	2 51	-0 40	2 30	1.23	

***** CANARD *****			***** WING *****						**FUSELAGE**		***MISC***	
%X/C	BP=3 375	BP=10.125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO	CP	
	CP	CP		CP	CP	CP	CP	CP	XLOC	CP	NO	CP
0 0	-8 5671	-3 8916	0 0	-0 6986	-0 5953	-2 0921	-2 8147	-7 2535	38.2	-0 8024	1	-1 3828
2 5	-8.5671	-3 7813	2 5	0 5918	0 2305	-1 9889	-2 4017	-5 6534	43.4	-0.7631	2	-0.6368
5 0	-2 3822	-4 0758	5 0	0 4369	0 0756	-1 9889	-2 4017	-4 5180	50.4	-0.5274	3	-2.1290
10 0	-0 0261	-4 1125	10 0	0 1273	-0 2340	-2 1437	-2 1954	-3 2277	56.1	-0.3310	4	-0 0399
15 0	-1 6091	-4 5175	15 0	-0 0792	-0 4921	-2 2470	-2 1437	-2 6600	62.5	-0.7238	5	-2 7258
25 0	-1 3882	-4 8857	24 0	-0.3889	-0 7502	-2 2470	-2 2470	-2 1437	69.7	-0.7631	6	-0.1891
35 0	-1 3515	-4 2230	33 0	-0.4875	-0.7860	-2.1290	-2.2782	-2 1290	76.9	-0.3703	7	-0.3062
50 0	-1 4619		54 0	-1.0844	-1.2336	-2.2782	-1.3828	-2 8751	83.4	-0 4488	8	-1.9688
56 0	-1 6460	-1 5355	65.0	-1.5095	-1.7060	-2.4916	-3 7880	-3 9845	89.4	-0.5274	9	-1.6172
65 0	-2 1613	-1.7196	78 5	-5.2022	-5.6736	-6.3022	-8.4236	-11 6448	95.4	-0.5274	10	0.8448
76 0	-0 6520	-4 9225	79 5*****	-118.2384	-113 0306	-120.8160	-128.6299	101.1	-0.4488	11	-0.5620	
79 0	-8.4934	-9 1930	80.5-87	8630	-66 7351	-75.1068	-52.9967	-24.3218	---BOTTOM---	12	-0.8498	
80 5	-33.1974	-52.9697	81.3-20	5476	162.7105	-16.2695	-7 2868	-18 2624	38.2	0.2583		
81 0	-9 8913	-33.2514	82.0 23	1687	15 1427	7.7016	-0 5899	-9.9709	43.4	0.2976		
82.0	-4.3370	-19 1665	84.0 -1	7857	-4 8153	-1.7060	-5.1077	-7.6588	50.4	0.0619		
84 0	-0.9087		87.0 0	2074	-6.3566	-5 9049	-3.6991	-6.7021	56.1	-0.0167		
87 0	-1.5200		89.0 -2	4485	-2 9280	-6.0296	-5.9656	-3.8234	62.5	-0 4095		
89 0	-2.8844	-5 0843	93 0 -1	4574	-1.6491	-2 9602	-1.0417	-7.3085	69.7	-0.5666		
93 0	-1 5488		96 0 -1	6491	-2,5764	-1.3294	-1.8090	-8.5555	76.9	-0.4488		
96 0	0.0226	0 8082	100.0 -1	3294	-1 4574	-0 8178	-1.1055	-1.1696	83.4	-0.2131		
100 0	1 4761	-1 5881	96.0 0	4611	0 2053	0.1733	0.4292	0 4611	89.4	0.0226		
96 0	0 5726	0 3369	84.0 0	3652	0 1733	0.3652	0 3972	0.3972	95.4	-0.0560		
84.0	1 4761	-1 5881	73.5 0	1094	0 1094	0.2586	0 2586	0.2586	101.1	-0.7631		
73 0	0.5726	0 3369	54 0 -0	3383	-0.3383	-0 1891	-0.0399	-0.0399				
50 0	0 8082	0 5333	33.0 -0	6368	-0 6368	-0.6368	-0.1891	-0.0399				
35.0	0.8206	0 5261	24 0 -0	6986	-0.7502	-0 8018	-0 4405	-0.0792				
25.0	0 4893	-2 0509	10.0 -0	5953	-0.5437	-0.9051	-0 7502	0.2586				
10 0	0.4893	0 3420	5 0 -0	5437	-0.4921	-1.1114	-0 8534	0.2586				
5.0	0.4893	0 3788	2 5 -0	5953	-0.5437	-1 3179	-1 0598	0.2586				
2 5	0.5261	0.2316										

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RUN SEQ	CLAERO =	O 5290	CDAERO =	O 1947	DCLC =	O O	DWLC =	O O	BASEPR =	8.1904
142 2	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUT	HGT	RN
ALPHA	0.06	2107 37	2077.42	29.95	160.73	O O	2 68	O O	O O	O O
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
O.O	O 53	O 19	-O 34	O.00	-O 00	O O1	O 52	-O 33	O.52	O 20

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
%X/C	BP=3 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP
	CP	CP		CP	CP	CP	CP	CP	XLOC		
0 0	-O 2198	-O 2116	0 0	O 5076	O 4387	O 1973	O 2088	-O.0555	38.2	-O.0831	1 -O 2943
2 5	-O 2198	-O 2198	2 5	-O 2164	-O.2854	-O 4807	-O.5152	-O 6761	43.4	-O 0656	2 O 0713
5 0	-O 2116	-O 2116	5.0	-O 2509	-O 3084	-O 4233	-O 5152	-O 5957	50 4	-O.0743	3 -O.3607
10 0	-O 2198	-O 2198	10.0	-O 2854	-O 3313	-O 4348	-O 4692	-O 4922	56 1	-O 0656	4 O 1045
15 0	-O 2280	-O 2198	15 0	-O 2969	-O 3084	-O 4233	-O.4003	-O 4578	62 5	-O.0743	5 -O 2943
25 0	-O 2280	-O 2116	24 0	-O 2739	-O 2854	-O 3658	-O 3543	-O 3428	69 7	-O 1268	6 O 0381
35 0	-O 2280	-O 2362	33 0	-O 2610	-O 2943	-O 3275	-O 3607	-O 3607	76 9	-O.1268	7 O 5034
50 0	-O 2198		54.0	-O 2943	-O 3607	-O 3607	O.3372	-O.3607	83.4	-O.1181	8 -O.3083
56 0	-O.2280	-O 2362	65.0	-O 3893	-O 4243	-O 4418	-O.4506	-O 4243	89 4	-O.1968	9 -O.0662
65 0	-O 2280	-O.2280	78.5	-O.6693	-O 6955	-O 7218	-O.7655	-O 8180	95.4	-O.2581	10 O.5817
76 0	-O 2444	-O 2280	79 5	-O 5310	-O 5369	-O 6375	-O.7027	-O.7973	101.1	-O.2056	11 O.2044
79 0	-O 2362	-O 2362	80.5	-O 4837	-O 4955	-O 6080	-O.7027	-O 2233	---BOTTOM---		12 -O.0591
80 5	-O 2114	-O.2055	81 3	-O 4955	54.0773	-O 6375	-O.6849	-O 8033	38.2	-O.0481	
81 0	-O 2055	-O.2055	82.0	-O.5074	-1.0282	-O 6612	-O 6967	-O.8210	43.4	-O.0218	
82 0	-O 2055	-O.1878	84.0	-O 4955	-O.5428	-O 6553	-O.6908	-O 8033	50.4	-O.0568	
84 0	-O 2174		87.0	-O 5133	-O 5310	-O.6553	-O.7086	-O 7973	56.1	O 0044	
87 0	-O 2055		89.0	-O.5860	-O.6074	-O.6857	-O 7569	-O 8494	62.5	O.0219	
89.0	-O 2056	-O.2056	93 0	-O.6145	-O.6074	-O.6857	-O 2514	-O.8637	69.7	O.0481	
93 0	-O.2056		96 0	-O 6145	-O 5789	-O.6786	-O 7569	-O.8565	76 9	O.0831	
96 0	-O 2056	-O 2143	100.0	-O.5433	-O.5219	-O 5860	-O.7426	-O.2442	83 4	O 1531	
100.0	-O 2056	-O 2143	96 0	O 3254	O.3752	O 3539	O.3325	O.2542	89 4	O 2056	
96 0	-O 2056	-O 2056	84 0	O 6387	O.6387	O.6316	O.6316	O 5176	95 4	O.1269	
84 0	-O.2056	-O 2143	73.5	O 5366	-O 1946	O.6030	O.6030	O.4701	101.1	-O.3193	
73 0	-O 2056	-O.2056	54 0	O.2042	O 2042	O 2707	O.2707	O 2042			
50 0	-O.1968	-O 2143	33.0	O 0713	O 0713	O.1045	O 1378	O 0713			
35.0	-O 2280	-O 2362	24 0	O 0134	O 0364	O 0364	O.1054	O.0249			
25 0	-O 2280	-O 2280	10 0	O 0249	O 0364	O 0709	O 1284	O.1045			
10 0	-O.2198	-O 2198	5.0	O 0594	O 0709	O 1284	O 1628	O 1710			
5 0	-O 2198	-O.2198	2 5	O 0939	O.1399	O 1743	O 2663	O.2375			
2.5	-O.2198	-O 2198									

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TABLE A11. (Continued)

RUN SEQ	142 4	CLAERO = 0.7048	CDAERO = 0.2423	DCLC = 0.0	DWLC = 0.0	BASEPR = 8.1904	PAGE 258				
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4 06	2107.23	2076 83	30.40	162 09	0 0	2 66	0.0	0 0	0 0	86.9151	0.100239E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0	0.70	0 24	-0.37	0 00	-0 00	0.01	0 71	-0.37	0.70	0.24	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **	
%X/C	BP=3.375	BP=10.125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO	CP	
	CP	CP		CP	CP	CP	CP	CP	XLCC			
0.0	-0.2437	-0 2276	0.0	-0.0529	-0.5510	-1 4228	-1.6379	-2 5324	38.2	-0.0773	1 -0 4404	
2 5	-0 2276	-0.2276	2.5	-0.2114	-0 8341	-1.0945	-1.2416	-1.5360	43.4	-0.0601	2 0 1816	
5 0	-0.2357	-0 2195	5.0	-0 4831	-0 6303	-0.8567	-0.9926	-1.1511	50.4	-0.0601	3 -0 5059	
10 0	-0 2195	-0 2114	10.0	-0.4831	-0 5397	-0.7322	-0 7661	-0.8227	56.1	-0.0687	4 0 1816	
15 0	-0 2276	-0.2195	15 0	-0 4378	-0 4718	-0 6303	-0.6529	-0.6755	62.5	-0.1032	5 -0 4404	
25.0	-0 2276	-0 2114	24.0	-0 3585	-0 4038	-0 5057	-0.4944	-0 5057	69.7	-0.1549	6 0 0179	
35 0	-0 2276	-0 2276	33 0	-0.3749	-0 4077	-0 4404	-0 4732	-0.4732	76.9	-0.1635	7 0 5091	
50.0	-0.2195		54 0	-0 3749	-0 3749	-0 4404	0.2798	-0.4404	83.4	-0.1463	8 -0.2625	
56 0	-0 2195	-0.2276	65.0	-0 4479	-0.4738	-0.4996	-0.5169	-0.5255	89.4	-0.2066	9 -0 0871	
65.0	-0 2276	-0.2195	78.5	-0 7324	-0.7324	-0 7496	-0 7927	-0 9047	95.4	-0.2497	10 0 5161	
76.0	-0.2437	-0 2357	79.5	-0.6207	-0.5799	-0 6616	-0 7140	-0 8423	101.1	-0.2066	11 0.2425	
79.0	-0 2276	-0.2276	80.5	-0 5799	-0.5391	-0.6499	-0.7199	-0.2010	---	---BOTTOM---	12 -0.0521	
80 5	-0.2243	-0 2068	81.3	-0 5683	47 6347	-0 6499	-0.7257	-0.8598	38.2	-0.0342		
81.0	-0 2243	-0.2068	82.0	-0.5799	-0.9881	-0 6616	-0 7082	-0 8598	43.4	0.0089		
82 0	-0 2126	-0 2010	84.0	-0.5741	-0.5624	-0.6616	-0.7082	-0.8831	50.4	-0.0256		
84 0	-0.2068		87.0	-0.5741	-0 6033	-0.6849	-0.7140	-0 8481	56.1	0.0347		
87.0	-0 2068		89 0	-0.6693	-0.6693	-0.7325	-0.7956	-0 9148	62.5	0.0951		
89 0	-0 1980	-0.1980	93.0	-0 7184	-0 6693	-0 7255	-0.2485	-0.9359	69.7	0.1295		
93 0	-0 2066		96.0	-0 6974	-0 6834	-0.7255	-0.7956	-0 9429	76 9	0.1554		
96 0	-0.2066	-0.1980	100.0	-0.6132	-0.5922	-0 5992	-0 7886	-0 2274	83.4	0.1985		
100 0	-0 1980	-0.1980	96.0	0.3407	0 3758	0.3477	0.3477	0.2215	89.4	0.2416		
96 0	-0 1980	-0 1980	84.0	0.7055	0 7125	0.6704	0 6073	0 5161	95.4	0.1468		
84 0	-0.1980	-0.1980	73.5	0 5745	-0.1785	0 6072	0 6072	0 4762	101.1	-0.3186		
73.0	-0 1980	-0 1980	54.0	0 2144	0 2798	0 3125	0.3125	0 1816				
50 0	-0 1807	-0 2066	33.0	0 1489	0 1489	0.1816	0.1816	0 1161				
35 0	-0.2276	-0.2276	24.0	0 1170	0.1509	0.1849	0.2415	0 1170				
25 0	-0 2276	-0.2195	10.0	0 2075	0.2302	0.2528	0 3208	0 2798				
10.0	-0.2276	-0 2195	5.0	0 2755	0 2981	0.3774	0.4113	0.3780				
5 0	-0.2357	-0 2195	2.5	0 3660	0.4113	0.4680	0.5019	0.4762				
2.5	-0 2276	-0.2276										

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TABLE A11 (Concluded)

RUN SEQ 142 6  
 ALPHA 11 98  
 BETA 0 0

CLAERO = 1.0886  
 PTOT 2107 09  
 CL 1.09

PROPLULSIVE CDAERO = 0 4181  
 VEL 161 60  
 CM 0 01

WING / CANARD DCLC = 0 0  
 H/C 2 65  
 CN 0 00

PRESSURES DWLC = 0 0  
 CMUC 0 0  
 CMUW 0 0  
 CMUT 0 0  
 CY 0 00  
 CNTR 1.14  
 CMTR -0.45

BASEPR = 8.1904  
 HGT 86 8355  
 CLTR 1.08

PAGE 260  
 RN  
 0.996939E+06  
 CDTR 0.42

***** CANARD *****			***** WING *****					**FUSELAGE**		***MISC***	
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	NO	CP
0 0	-0 2766	-0 2684	0.0	-2 8887	-5.6604	-6 0252	-2 3412	-1 4058	38.2	-0.1019	1 -0.6812
2 5	-0 2603	-0 2684	2 5	-0 2310	-2 1700	-4.4397	-2.4210	-1 4400	43 4	-0.0932	2 0 3413
5 0	-0 2684	-0.2684	5 0	-1 1663	-1 5769	-2 6832	-2 3069	-1 4172	50 4	-0 1279	3 -1 4068
10 0	-0.2603	-0 2603	10.0	-0 9496	-1 2119	-1 3373	-2.3866	-1 3830	56 1	-0 1627	4 0 3083
15 0	-0.2603	-0 2603	15 0	-0 8241	-0 9724	-1 1663	-2.2042	-1 3487	62 5	-0.2234	5 -1.1429
25 0	-0 2684	-0 2684	24 0	-0 6758	-0 7556	-0 9610	-1.7023	-1 2917	69 7	-0 3016	6 0 0774
35 0	-0 2684	-0 2684	33.0	-0.5822	-0 6812	-0 7801	-1.2089	-1 2749	76 9	-0 2669	7 0 5063
50 0	-0.2766		54 0	-0 5163	-0.5493	-0 5822	0.1764	-1 1100	83.4	-0.2061	8 -0 2923
56 0	-0 2603	-0 2603	65 0	-0 5534	-0 5795	-0 6055	-0 5447	-1 0136	89 4	-0.2755	9 -0.0096
65 0	-0 2766	-0 2603	78 5	-0 8139	-0 8052	-0.8226	-0 8573	-0.9962	95.4	-0.2321	10 0 5274
76 0	-0 2766	-0 2684	79 5	-0 6496	-0 6437	-0 7612	-0.8728	-1 0020	101.1	-0.1627	11 0.0964
79 0	-0 2684	-0 2684	80 5	-0 6261	-0 6144	-0 7612	-0 8376	-0 2502	---	BOTTOM---	12 -0.0874
80 5	-0 2678	-0 2561	81 3	-0 6085	43 2572	-0.7671	-0 8435	-0.9962	38.2	0.0197	
81 0	-0 2502	-0 2502	82 0	-0 6026	-1.0197	-0 7201	-0 8670	-0.9844	43 4	0 0544	
82 0	-0 2443	-0 2619	84 0	-0 6203	-0 6261	-0 7318	-0.8611	-0.9727	50 4	0 0110	
84 0	-0 2443		87.0	-0 6672	-0 6672	-0 7436	-0 8258	-0 9727	56.1	0.1326	
87 0	-0 2561		89 0	-0 7304	-0 7163	-0 8152	-0.9353	-1 0060	62 5	0 2107	
89 0	-0 2408	-0 2495	93 0	-0 7799	-0 7304	-0 8152	-0.3135	-1.0060	69 7	0 2802	
93 0	-0.2495		96.0	-0 7658	-0 6951	-0 8081	-0.9212	-0.9777	76 9	0 2715	
96 0	-0 2495	-0 2495	100 0	-0 6597	-0 5962	-0 6527	-0 8717	-0 2852	83 4	0.2889	
100 0	-0 2495	-0 2582	96 0	0.3367	0 4356	0 3720	0 3508	0 1883	89 4	0.3062	
96 0	-0 2495	-0 2582	84 0	0 7112	0 7606	0 6829	0.6264	0.4285	95.4	0.1673	
84 0	-0 2495	-0 2582	73.5	0 6051	-0 2524	0 6381	0 6051	0 4402	101 1	-0.3190	
73 0	-0 2495	-0.2582	54.0	0 3413	0.3413	0.3742	0.3413	0 2093			
50 0	-0 2495	-0.2582	33 0	0 3413	0 3413	0 3413	0 3413	0 1764			
35 0	-0 2684	-0 2684	24 0	0 3393	0 3507	0 3735	0 3964	0 2481			
25 0	-0 2684	-0.2684	10 0	0 4990	0 4876	0 5104	0 5333	0 4072			
10 0	-0 2603	-0.2603	5 0	0.6131	0 5789	0 5789	0 6017	0 4732			
5 0	-0.2684	-0 2684	2.5	0.6929	0 5789	0 5447	0 5447	0 4732			
2 5	-0 2603	-0 2684									

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TABLE A12. RUN 143, BW6V, DELF=45, CMU=0 5, (BN/B)=1

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 261				
143 1	CLAERO =	0.2820	CDAERO =	0 6253	DCLC =	0 0	DWLC =	0.4420	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0 05	2107.23	2079.31	27.91	155.49	0 0	2 66	0 0	0.510	0 510	86.9365	0.957290E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR
0 0	0.72	0.37	-0 80	0 00	-0 01	0 06	0.35	-0 53		0.35	0.54

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
	BP=3 375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0.0	-0.4254	-0.4166	0.0	0 1611	-0.5170	-1.6390	-2.0089	-3 4267	38.2	-0.0987	1	-0.7118
2.5	-0.4254	-0 4078	2 5	-0.4061	-0 8129	-1.4294	-1.7130	-1 9596	43.4	-0.0893	2	0 2865
5 0	-0.4166	-0 4166	5 0	-0 4924	-0.6897	-0 9979	-1.2322	-1.6514	50.4	-0.1081	3	-0 9614
10.0	-0 4254	-0 4166	10.0	-0.5540	-0.6650	-0 8992	-1.0349	-1.1828	56.1	-0.1174	4	0 2152
15.0	-0.4254	-0.4078	15 0	-0.5047	-0.6280	-0.8252	-0.9362	-1.0595	62.5	-0.1644	5	-0 9257
25 0	-0.4254	-0.4166	24 0	-0 5294	-0.6157	-0.7636	-0 8006	-0 9239	69 7	-0.2676	6	-0 0701
35.0	-0 4078	-0.4166	33 0	-0.5692	-0 6761	-0 7831	-0.8544	-0.9257	76 9	-0.3052	7	0 2510
50 0	-0.4254		54 0	-0 8188	-0 9257	-1.0683	0.2152	-1.2466	83.4	-0.3239	8	-0 9102
56.0	-0.4342	-0 4254	65 0	-1.2344	-1.3564	-1 4221	-1.5536	-1 7319	89 4	-0.4272	9	-0.5740
65.0	-0.4254	-0 4254	78 5	-3.2619	-3.5059	-3.1211	-4.0505	-5.5709	95 4	-0.4178	10	0 8697
76.0	-0.4430	-0.4254	79.5	-15.8768	-18.2134	-9.5719	-19.1087	-37.0007	101.1	-0 3333	11	0 0294
79 0	-0.4254	-0 4254	80.5	-17 1723	-17.1404	-9.1022	-16.6958	-0.0735	---	---BOTTOM---	12	-0.2914
80.5	-0 4227	-0 4037	81.3	-17 6928	46.1040	-10.7151	-17.3944	-7.4514	38.2	-0.0330		
81 0	-0 4100	-0 4037	82.0	-16.9183	-16.3977	-11.2865	-15.0643	-4.3593	43.4	-0.0142		
82 0	-0 4037	-0.4164	84.0	-7.6355	-8 6261	-11.5341	-9.2419	-3.9021	50.4	-0.0611		
84 0	-0.4164		87 0	0 0725	-2 1053	-7.4324	-3.5718	-3 2734	56.1	0.0515		
87 0	-0.4100		89 0	-0 3296	-0.4365	-3 2476	-1 5060	-2.9495	62.5	0.1078		
89 0	-0 3615	-0 4084	93 0	-0.6733	-0 4824	-0.3296	-0.4824	-3.6908	69.7	0.1735		
93 0	-0 3990		96 0	-0.8338	-0 4977	0 3120	-0 8109	-3.5532	76 9	0.2392		
96 0	-0.3990	-0 4084	100 0	0 0523	0.1975	0 0600	-0.0240	-0.4289	83 4	0.3143		
100.0	-0.3990	-0 4084	96.0	0 5870	0 7169	0 6787	0.6176	0.2204	89.4	0.3613		
96 0	-0 3990	-0 4084	84 0	0 6634	0.6787	0 7474	0.6176	0 0982	95 4	0.2111		
84 0	-0 3990	-0 4084	73.5	0 6073	-0 4266	0 6073	0 5717	0 1795	101 1	-0.3427		
73.0	-0.3990	-0.4084	54 0	0 3578	0 4291	0 4291	0.3934	0.1082				
50 0	-0 4084	-0 3990	33 0	0.2152	0 2508	0.2865	0 2865	0 1082				
35.0	-0.4166	-0 4254	24 0	0.1734	0 1981	0 2598	0 2844	0 1241				
25 0	-0 4342	-0 4254	10 0	0.1858	0 2228	0 2844	0 3830	0 2865				
10 0	-0 4166	-0 4254	5 0	0 2598	0.2721	0 3830	0 4200	0 3934				
5 0	-0 4254	-0.4254	2.5	0.3707	0.3584	0 4693	0.5064	0 4291				
2 5	-0 4166	-0.4342										

Force Data -  
Use Run 313

A-243

TABLE A12 (Continued)

RUN SEQ 143 3  
 ALPHA 4 01  
 BETA 0.0

CLAERO = 0 2949  
 PTOT 2107 16  
 CL 0 76

PROPLULSIVE CDAERO = 0 6624  
 VEL 157.44  
 CROLL -0.01

WING / CANARD DCLC = 0 0  
 YAW 0.0  
 CN -0 02

PRESSURES DWLC = 0.4681  
 CMUC 0.0  
 CMY 0.07

BASEPR = 8.1904  
 HGT 87 0371  
 CLTR 0.37

PAGE 263  
 RN  
 0.967873E+06  
 CDTR  
 0.57

\*\*\*\*\* CANARD \*\*\*\*\*

%X/C	BP=3 375 CP	BP=10 125 CP
0 0	-0.4166	-0.3822
2.5	-0 4166	-0 3564
5 0	-0 4080	-0 3479
10 0	-0 3994	-0 3651
15 0	-0.3994	-0.3736
25 0	-0 3994	-0 3736
35 0	-0 3994	-0 4080
50 0	-0.3908	
56 0	-0 3908	-0 4080
65.0	-0 4080	-0 4080
76 0	-0 4166	-0 4166
79.0	-0 4080	-0 4251
80.5	-0.4006	-0 3820
81 0	-0 3882	-0 3820
82 0	-0 3882	-0 3944
84 0	-0.3696	
87 0	-0.3882	
89 0	-0.3101	-0.3468
93 0	-0 3284	
96 0	-0 3284	-0 3468
100 0	-0 3376	-0 3743
96 0	-0 3468	-0 3651
84 0	-0 3376	-0.3743
73 0	-0 3468	-0 3651
50 0	-0 3376	-0 3651.
35 0	-0 4080	-0 4251
25 0	-0.3994	-0 4251
10 0	-0 4166	-0 4338
5 0	-0 4080	-0 3736
2 5	-0 3908	-0 3822

\*\*\*\*\* WING \*\*\*\*\*

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0 8775	-1 8044	-3 9953	-5 5000	-2.4183
2.5	-0 3840	-1.2868	-2.0692	-2 5987	-2 2136
5.0	-0 7692	-0 9257	-1.4553	-1.8525	-2 2618
10.0	-0 7451	-0.8414	-1.2025	-1.4312	-2.1053
15 0	-0 7090	-0.7331	-1.0219	-1 2506	-2 0090
24 0	-0 6247	-0.6608	-0.8896	-1 0701	-1.7201
33.0	-0 7052	-0.7749	-0 7749	-0.9837	-1 0533
54 0	-0 8793	-0.9837	-0 9489	0 3042	-0 9489
65 0	-1 2356	-1.3731	-1 4556	-1.5930	-1.9779
78.5	-3.0866	-3 4259	-3 0317	-3.9114	-5.6158
79 5-11	7443	-3 9897	-7.6409	-12 9408	-32.1566
80.5	-9 4012	-3.7603	-7 0333	-12.4511	-0.1217
81.3	-6 4692	38 3652	-6.9092	-12.8415	-13 5543
82.0	-2 3222	-3 5186	-7 5478	-12.8166	-7 7958
84.0	-2 5578	-7 4859	-6.8969	-11.3291	-3 4567
87.0	-3 6365	-3 5249	-7 4114	-8.7503	-3 6365
89.0	-0 0959	-0 6329	-6.4199	-7.2404	-3 0940
93.0	-0 3271	-0 5658	-2.9446	-0.4763	-3 8397
96 0	-0.4912	-0 4688	-0.5136	-0.5956	-3 6756
100.0	0.1427	0 1651	-0.0288	0.0085	-0 4017
96 0	0 6200	0 6796	0 6796	0 6349	0.2247
84 0	0 7318	0 7095	0 8213	0.6424	0 0606
73 5	0 6522	-0 4268	0 6522	0 5826	0.1649
54 0	0 4086	0 4434	0 4434	0 4086	0.1301
33.0	0.3042	0 3042	0 3042	0.3042	0.0953
24 0	0 2660	0 2660	0 3142	0 3382	0.1577
10 0	0.3744	0 3623	0 3984	0 4586	0 3042
5 0	0 4225	0.4346	0 4827	0 5429	0 4434
2.5	0 5309	0 5068	0 5429	0 4947	0 4434

\*\*FUSELAGE\*\*

XLOC	CP	NO	CP
38.2	-0.1177	1	-0.8097
43.4	-0 0993	2	0 3042
50 4	-0.1268	3	-1 0185
56.1	-0.1452	4	0.2693
62.5	-0.2093	5	-1.3318
69.7	-0.3284	6	-0 1135
76.9	-0.3651	7	0 2173
83.4	-0 3743	8	-0.9759
89 4	-0.5209	9	-0.6702
95 4	-0 4567	10	0 8661
101 1	-0.3743	11	-0 0064
		12	-0 3495

---TOP---

---BOTTOM---

Force Data -  
 Use Run 313

A-244

TABLE A12 (Concluded)

RUN SEQ		PROPLULSIVE WING / CANARD PRESSURES										PAGE 265	
143	5	CLAERO =	0 4314	CDAERO =	0 7349	DCLC =	0 0	DWLC =	0.4717	BASEPR =	8.1904		
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
11 97		2107.16	2079 58	27 57	154 67	0 0	2 66	0 0	0.496	0 496	87.1425	0.949446E+06	
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0 0		0 90	0 58	-0.82	0 02	-0.02	0 05	0.63	-0 54	0.50	0.67		
***** CANARD *****			***** WING *****					**FUSELAGE**					
		BP=3 375	BP=10 125	BP = 2		BP = 6	BP =12	BP =16	BP = 22	----TOP----		**MISC.**	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	-0 5127	-0.4949	0.0	-4.8901	-9.3958	-5 3768	-3.5421	-2.6812	38.2	-0 1203	1	-1.1284	
2.5	-0.4949	-0 4771	2 5	-0.4469	-4 0788	-5 4145	-3 7670	-2.6561	43 4	-0.1203	2	0 4957	
5 0	-0 4949	-0 4771	5 0	-1 9197	-2.3690	-5 2647	-3 8544	-2 7435	50.4	-0.1868	3	-4.1601	
10 0	-0 4860	-0 4682	10 0	-1.5203	-1.8448	-4 3911	-4.1412	-2.8556	56.1	-0.2438	4	0 4235	
15 0	-0.4949	-0 4593	15 0	-1.3081	-1 5452	-3 7920	-4.2163	-2.8433	62.5	-0.3483	5	-2.8607	
25 0	-0 4860	-0.4593	24 0	-1.1583	-1 2707	-1.1833	-3.8667	-2.9183	69.7	-0.4813	6	-0.0818	
35 0	-0 4593	-0 4326	33 0	-1 0563	-1 1645	-0.8758	-1.9946	-3.3300	76.9	-0 4813	7	0.0885	
50.0	-0 4593		54 0	-1 1284	-1 2728	-1.2728	0.2430	-4 5209	83 4	-0.4053	8	-1.3807	
56 0	-0 4682	-0 4237	65 0	-1.5455	-1 7071	-1.6880	-1.4600	-4.8140	89 4	-0.4813	9	-0.9863	
65.0	-0 4415	-0 4237	78 5	-3 7024	-3 7593	-3 8355	-4 7665	-6.3629	95.4	-0.3958	10	0 9004	
76 0	-0 4415	-0 4326	79.5	-17 8055	-20 1066	-11 3973	-28.7448	-38.4057	101.1	-0.3198	11	-0 1899	
79 0	-0 4415	-0 4148	80.5	-17 3685	-18 7245	-11 9116	-23.4743	-0.0531	---BOTTOM---		12	-0.5997	
80 5	-0 4838	-0.4645	81.3	-18.7633	34 3011	-12 8052	-19.1811	-8.6016	38.2	0.0128			
81.0	-0 4709	-0 4581	82 0	-17.4135	-17 4969	-12.7279	-16.4492	-6 8148	43 4	0.0508			
82 0	-0.4645	-0.4517	84 0	-6 3070	-6 3712	-13 4736	-3.6589	-5 9600	50.4	-0.0063			
84 0	-0 4517		87 0	0 7631	-1 6278	-6.6991	-2 0135	-4 7131	56.1	0.1648			
87 0	-0.4645		89 0	-0 4992	-1.3885	-2.2390	-2.5715	-3 9788	62.5	0.2693			
89 0	-0 4338	-0.4528	93 0	-1.0018	-0 8781	0.2432	-0.5533	-4.3965	69.7	0.3738			
93 0	-0.4623		96 0	-0 9863	-0.6771	-0.0739	-1.3189	-3.9944	76 9	0.4118			
96 0	-0.4528	-0 4528	100 0	-0 0198	0.1581	0 0266	-0.0430	-0.4296	83 4	0.4593			
100 0	-0 4528	-0 4528	96 0	0.5602	0 7690	0.7148	0.5989	0.1349	89 4	0.4688			
96 0	-0.4433	-0 4433	84 0	0 7535	0.8849	0 7612	0.6530	0 0266	95 4	0 2978			
84 0	-0 4528	-0 4528	73.5	0.6761	-0 4427	0 6761	0 6400	0 1348	101 1	-0.2723			
73 0	-0 4433	-0 4433	54 0	0 4595	0 5318	0 5318	0.4595	0.0987					
50 0	-0.4433	-0.4528	33 0	0 4595	0 4595	0.4957	0 4595	0.1708					
35 0	-0 4415	-0 4237	24 0	0 4393	0 5142	0.5142	0.5266	0 2770					
25 0	-0.4415	-0 4148	10 0	0 6015	0 5766	0 5891	0 5891	0.4595					
10 0	-0 4504	-0 4237	5 0	0 7014	0 6265	0 5891	0.5516	0.3874					
5 0	-0 4860	-0 4593	2.5	0.7388	0 4767	0.4018	0.2770	0.2069					
2 5	-0 4860	-0 4593											

Force Data -  
Use Run 313

A-245

RUN SEQ	PROPLULSIVE										WING / CANARD		PRESSURES		PAGE 266	
144 1	CLAERO =	0 8342	CDAERO =	0 8265	DCLC =	0 0	DWLC =	0.8711	BASEPR =		8.1904					
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN					
0 07	2107 86	2094 08	13.79	109 08	0 0	2 66	0 0	1.005	1 005	86 9286	0.673305E+06					
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR						
0 0	1 71	0 33	-1 38	0 02	-0 01	0 07	0 96	-0.82	0 96	0 66						

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
	BP=3 375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XL0C	CP	
0 0	-0 4949	-0 4949	0 0	0.1772	-0 4968	-1 6949	-3 1925	-4 3408	38.2	-0.1013	1 -0.8757
2 5	-0 4949	-0 4949	2 5	-0 4718	-1 0459	-1 3954	-1 8197	-2 6933	43.4	-0.1013	2 0 2069
5 0	-0 4949	-0 4949	5 0	-0 5717	-0 7713	-1.1458	-1 4703	-1 8946	50 4	-0.1393	3 -1.0923
10 0	-0 4949	-0 4771	10.0	-0 5717	-0 7214	-0 9960	-1 1708	-1 3704	56 1	-0.1203	4 0.2069
15 0	-0 4949	-0 4593	15 0	-0 5966	-0.7464	-0.9710	-1 0459	-1 2456	62.5	-0.1773	5 -1.0923
25 0	-0 4771	-0 4771	24.0	-0 5966	-0 6965	-0 8463	-0 9461	-1 0958	69 7	-0.2913	6 -0.0818
35 0	-0 4949	-0 4771	33.0	-0 6592	-0.7314	-0 9479	-1 0201	-1 0923	76.9	-0.3483	7 0.1658
50 0	-0 4771	-0 4771	54 0	-0 9479	-1.0923	-1.3088	-0.2983	-1 5253	83.4	-0.3673	8 -1 0713
56 0	-0.5127	-0 4949	65.0	-1 4694	-1 6594	-1.7735	-1.9065	-2.0585	89.4	-0.4813	9 -0 6848
65 0	-0 4771	-0 4771	78.5	-4.4527	-4 9278	-5 1938	-5.7829	-6 7138	95.4	-0.4433	10 0.8926
76 0	-0.4949	-0.4949	79.5	-46.6094	-56 5970	-46.5456	-60.3503	-62 0467	101 1	-0.3673	11 -0.0971
79 0	-0.4771	-0 4771	80 5	-44 6685	-42.8561	-40.5809	-31.6609	0.0625	---	---BOTTOM---	12 -0 3600
80 5	-0 5031	-0 5287	81.3	-27 5471	68.0077	-28 3188	-7 3927	-9.8992	38.2	-0.0443	
81 0	-0 5031	-0 5159	82 0	-4 7576	7.1964	-17.1098	-1.9427	-6 4027	43.4	-0 0063	
82 0	-0 5159	-0.4902	84 0	0 9366	-3.0481	3.3788	-3.5494	-4 9119	50.4	-0.0823	
84 0	-0 4902		87 0	-0 5159	-3 8836	-2 8296	-2.7525	-3.7808	56.1	0.0507	
87 0	-0 5031		89.0	-2.1383	-2 5868	-3 4992	-3 9941	-3 1281	62.5	0.1077	
89 0	-0 4243	-0 4623	93 0	-1 5198	-1 3961	-1 8909	-0 6074	-4 5044	69 7	0.1648	
93 0	-0 4813		96 0	-1 5043	-1 4734	-0 9786	-1 1487	-4 6900	76.9	0.2408	
96 0	-0 4623	-0 4623	100 0	-0 5610	-0 4837	-0 3445	-0 6538	-0 4992	83.4	0.3168	
100 0	-0.4813	-0 4623	96 0	0 5679	0 6761	0 6452	0.5988	0.1194	89 4	0 3548	
96 0	-0 4623	-0 4623	84.0	0.6297	0 7225	0.7380	0.5833	0 0420	95.4	0.1458	
84 0	-0 4813	-0 4623	73 5	0 5678	-0 4427	0 6400	0 5678	0.2069	101 1	-0.5003	
73 0	-0 4623	-0 4623	54 0	0 3513	0 3513	0 4234	0 3513	0 0626			
50 0	-0 4623	-0 4623	33 0	0 2069	0 2069	0.2069	0 2069	0 0626			
35 0	-0 4771	-0 4949	24.0	0 1772	0 2021	0 2520	0.3020	0 1272			
25 0	-0.4949	-0 4771	10 0	0 2021	0 2271	0.3020	0 4018	0.2791			
10 0	-0 4949	-0 4771	5.0	0 2021	0 3269	0 4018	0.4517	0 3513			
5 0	-0 4771	-0 4949	2.5	0 2770	0 3519	0.4767	0.4767	0.4234			
2 5	-0.4771	-0 4949									

Force Data -  
Use Run 312

A-246



TABLE A13 (Continued)

RUN SEQ	144 3	CLAERO =	0 8837	CDAERO =	0 9060	DCLC =	0 0	DWLC =	0 9225	BASEPR =	8.1904	PAGE	268
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
4.04	2108 08	2094 52	13.56	108 13	0 0	2.66	0 0	1.026	1.026	86.9895	0.668554E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
0.0	1.81	0 46	-1.41	0.02	-0.02	0 08	1 07	-0 84	1.02	0.74			

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10 125
%X/C	CP	CP
0 0	-0 5178	-0 5178
2 5	-0 4997	-0.4816
5 0	-0.4997	-0.5178
10 0	-0 4997	-0.4997
15 0	-0.4997	-0 5178
25 0	-0 5178	-0.5178
35 0	-0 5178	-0.4997
50 0	-0.5178	
56 0	-0.4997	-0 5178
65 0	-0.5178	-0.5178
76 0	-0.5360	-0.4997
79 0	-0.5178	-0.5178
80 5	-0.4739	-0 4608
81 0	-0.4608	-0.4739
82.0	-0.4608	-0 4347
84 0	-0.4608	
87.0	-0.4608	
89.0	-0 4268	-0.4847
93 0	-0.4654	
96.0	-0.4654	-0 4847
100.0	-0 4654	-0 4847
96.0	-0 4654	-0.4847
84.0	-0 4654	-0.4847
73 0	-0.4654	-0 4847
50 0	-0.4654	-0 4847
35 0	-0 4997	-0.4997
25 0	-0 5178	-0 5178
10.0	-0.4997	-0 5178
5 0	-0.5178	-0.5178
2.5	-0.4997	-0.5178

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0 0	-0 9765	-2.3723	-5 1891	-4.8592	-2.4992
2 5	-0.5198	-1.7125	-2.4230	-4.1486	-2.3469
5 0	-0.9258	-1.2811	-1.8901	-4.0725	-2.3723
10 0	-0 8496	-1.1035	-1 5349	-1 3319	-2 2707
15 0	-0.8496	-1.0019	-1 3572	-1.1796	-2 1693
24 0	-0 7736	-0.9258	-1 1542	-1.1288	-2.0678
33 0	-0.8317	-0.9785	-1 1252	-1.2719	-2.0057
54 0	-1 1252	-1.1986	-1.4187	-0.2447	-1.5655
65 0	-1.5859	-1.7984	-1.9143	-2.0689	-2.7257
78.5	-4.5996	-5.1405	-5 3917	-6.1644	-7.3817
79.5-47	2570	-57 2673	-44.5259	-60.5995	-63.2657
80.5-45	6496	-45 2057	-40 4358	-35.3259	0.0880
81 3-29	2626	63.2849	-31 9938	-10.4970	-10.9940
82 0	-7 0344	7.2754	-22 4804	-2.1858	-7.3477
84 0	1 4079	-2 8915	3.7601	-3.4142	-5.7403
87 0	-0 4347	-3 9108	-2.4264	-2.7999	-4.6164
89 0	-2.1573	-2.6133	-3 5094	-4.2798	-3.9339
93 0	-1.4970	-1.4341	-2 0001	-0.6008	-5 4274
96 0	-1.4498	-1.4655	-1.1197	-1.3240	-5.5375
100 0	-0 5065	-0 4908	-0.3178	-0 6952	-0 4908
96 0	0.6255	0.7041	0.6569	0.5783	0.0438
84 0	0 7198	0.8142	0.7827	0.6412	-0.0191
73 5	0.6359	-0.4648	0 6359	0.5625	0.0489
54 0	0 4158	0 4158	0 4158	0.3424	-0.0245
33 0	0.2690	0.3424	0 3424	0 2690	-0 0245
24 0	0.2669	0 2923	0.3176	0.3938	0 0892
10 0	0.3430	0 3430	0.4191	0.4953	0 3424
5 0	0 4191	0 4191	0.4699	0 5714	0 4158
2.5	0 4953	0 5206	0.5206	0.5206	0 3424

\*\*FUSELAGE\*\*

----	TOP----	**MISC.**
XLOC	CP	NO. CP
38.2	-0.1177	1 -0.9785
43.4	-0.1177	2 0.3424
50.4	-0.1370	3 -1.3454
56.1	-0.1950	4 0.1956
62.5	-0.2529	5 -1.8590
69.7	-0 3688	6 -0.2447
76.9	-0.4075	7 0.1538
83.4	-0.4268	8 -1.1983
89.4	-0.5234	9 -0.8209
95.4	-0.4847	10 0.8928
101.1	-0.4075	11 -0.0977
---	BOTTOM---	12 -0.3964
38.2	-0.0404	
43.4	-0.0018	
50.4	-0.0790	
56.1	0.0562	
62.5	0.1528	
69.7	0.2300	
76.9	0.2880	
83.4	0.3460	
89.4	0.4039	
95.4	0.1914	
101.1	-0.4847	

Force Data -  
Use Run 312

A-247

TABLE A13 (Concluded)

RUN SEQ	CLAERO =	1 0612	CDAERO =	1 0923	DCLC =	0 0	DWLC =	1 0102	BASEPR =	8.1904	PAGE	270
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
144 5	2107 86	2094.76	13 11	106 28	0 0	2 66	0 0	1.062	1.062	86.9901	0.657723E+06	
0 0	2 07	0 76	-1 46	0 03	-0 02	0.06	1.39	-0.87	1.22	0 94		
***** CANARD *****												
%X/C	BP=3 375	CP	BP=10 125	CP	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	***** WING *****	
0 0	-0 5551	-0 5364	0 0	-5 3803	-8 4255	-4 9865	-3 8051	-3 0437	38.2	-0 1799	**FUSELAGE**	
2 5	-0 5364	-0 5551	2 5	-0 5234	-7 4277	-5 3803	-4 0676	-3 0963	43 4	-0 1999	-----TOP----	
5 0	-0 5364	-0 5364	5 0	-2.0461	-2 9388	-5.2753	-4 3302	-3 2801	50.4	-0.2598	**MISC.***	
10 0	-0 5739	-0 5364	10 0	-1 7048	-1 5998	-6 1154	-4 4352	-3 3325	56.1	-0 3398	NO CP	
15 0	-0 5551	-0 5177	15 0	-1 4948	-1.5473	-4 5139	-5 0916	-3 2275	62.5	-0 4397	1 -1 2782	
25 0	-0 5364	-0 5177	24 0	-1 2585	-1 3373	-1 3635	-4 7765	-3 5163	69.7	-0 5796	2 0 3920	
35 0	-0 5551	-0 5739	33 0	-1 2022	-1.2782	-1 0504	-2 8724	-3 6593	76 9	-0 5796	3 -4 8462	
50 0	-0 5551	-0 5551	54 0	-1 3540	-1 4300	-1 3540	-0 4431	-5 3016	83 4	-0 5197	4 0 3920	
56 0	-0 5364	-0 5551	65 0	-1 7788	-1 9986	-1 9387	-1 6589	-5 7361	89 4	-0 6196	5 -3 4038	
65 0	-0 5551	-0 5551	78.5	-4 7767	-5 4763	-5 7561	-6 1558	-8 0342	95.4	-0 5396	6 -0 2154	
76 0	-0 5739	-0 5364	79 5	-49 7803	-58 6757	-44 4672	-63 4350	-59 6766	101.1	-0 4597	7 -0 0579	
79 0	-0 5551	-0 5551	80 5	-47 9419	-48 7123	-41 2625	-37 6261	-0 0282	---	BOTTOM---	8 -1 5869	
80 5	-0 5689	-0 5419	81 3	-29 3928	60 2288	-35 2872	-12 1550	-12 6959	38 2	-0 0000	9 -1 1802	
81 0	-0 5149	-0 5419	82 0	-6 8014	6 9479	-26 9456	-2 2995	-9 0457	43 4	0 0200	10 0 8855	
82 0	-0 5419	-0 5284	84 0	1 5536	-2 8538	2 3783	-3 9218	-7 7207	50.4	-0 0600	11 -0 2369	
84 0	-0 5419		87 0	-0 4879	-4 0841	-2 0426	-3 5162	-5 8281	56.1	0 1399	12 -0 7085	
87 0	-0 5554		89 0	-2 2862	-2 7580	-3 5549	-4 8236	-4 7748	62.5	0 2598		
89 0	-0 4597	-0 5396	93 0	-1 5543	-1 6031	-2 2212	-0 6597	-5 8971	69 7	0 3597		
93 0	-0 5396		96 0	-1 4730	-1 5706	-1 2941	-1 5869	-5 6857	76 9	0 3997		
96 0	-0 5396	-0 5197	100 0	-0 1555	-0 3344	-0 2369	-0 6923	-0 5297	83 4	0 4197		
100 0	-0 5396	-0 5396	96 0	0 5927	0 7878	0 6740	0 5764	-0 0092	89 4	0 4397		
96 0	-0 5197	-0 5396	84 0	0 7065	0 9017	0 8529	0 6903	0 0397	95 4	0 1799		
84 0	-0 5396	-0 5396	73 5	0 6197	-0 5190	0 6956	0 6197	0 0124	101.1	-0 5396		
73 0	-0 5197	-0 5396	54 0	0 3920	0 5438	0 4679	0 3920	-0 0635			Force Data -	
50 0	-0 5197	-0 5396	33 0	0 3920	0 4679	0 3920	0 3920	0 0124			Use Run 312	
35 0	-0 5551	-0 5739	24 0	0 4480	0 4742	0 4742	0 5267	0 1854				
25 0	-0 5551	-0 5551	10 0	0 6055	0 5530	0 5530	0 5267	0 3161				
10 0	-0 5551	-0 5551	5 0	0 6580	0 5792	0 5005	0 4480	0 3161				
5 0	-0 5551	-0 5364	2 5	0 6842	0 4480	0 2642	0 1854	0 0124				
2 5	-0 5364	-0 5551										

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RUN SEQ	145 1	CLAERO =	1.8098	CDAERO =	1 1250	DCLC =	0 0	DWLC =	1.7854	BASEPR =	8.1904
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0 04	2107 72	2101 06	6 67	75 69	0 0	2 67	0 0	2 061	2 061	87.2267	0.470101E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0 0	3 60	0.10	-2 54	0 03	0 00	0 02	2 03	-1 37	2.03	0 79	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **	
%X/C	BP=3 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO	CP	
	CP	CP		CP	CP	CP	CP	CP	XLOC			
0 0	-0.6671	-0 6671	0 0	-0 1457	-1 1780	-2 6232	-3 8619	-3 2941	38.2	-0 1890	1 -0 9504	
2.5	-0 6304	-0 6304	2 5	-0 7135	-1.2812	-1 8490	-2.1070	-3 2426	43 4	-0.1890	2 0 2435	
5 0	-0.6671	-0 6304	5 0	-0 7135	-0 9715	-1 5393	-1 7974	-2.8812	50 4	-0.2283	3 -1.2490	
10 0	-0 6671	-0 6304	10.0	-0 8168	-0.9715	-1 3844	-1.4877	-1 8490	56.1	-0.2676	4 0 2435	
15 0	-0 5935	-0 6671	15 0	-0.7650	-0 9200	-1.2296	-1.3328	-1.8490	62.5	-0 3855	5 -1.2490	
25 0	-0 6304	-0 6671	24 0	-0 7650	-0 9200	-1.2296	-1.2296	-1.4877	69.7	-0.4248	6 -0.2042	
35 0	-0.6671	-0 6671	33 0	-0 8012	-0 9504	-1 0996	-1.2490	-1 2490	76.9	-0.5033	7 -0.0015	
50 0	-0 6671		54.0	-1.2490	-1 2490	-1 5474	-1.8459	-1.8459	83.4	-0.5033	8 -1.4084	
56 0	-0 6671	-0 6671	65.0	-1.8393	-2 0357	-2.1536	-2.3108	-2.5072	89.4	-0.6605	9 -0.8969	
65 0	-0 6671	-0.7040	78 5	-5.8864	-6 3972	-6.8294	-7.3402	-8.2439	95.4	-0.6212	10 0.8299	
76 0	-0.7040	-0 7040	79 5	*****	-144.3701	-132 3830	-128 5018	-84.8605	101.1	-0.5426	11 -0.1294	
79 0	-0 7040	-0 7040	80 5	-11 3429	-13.0173	-15.4889	-24 9506	0.1392	---	---BOTTOM---	12 -0.4172	
80 5	-0 6581	-0 6050	81.3	0.8835	131 3587	-4 0602	-5.4955	-15.3825	38.2	-0.1104		
81 0	-0.6316	-0 5518	82 0	7 8737	-6 7713	6 0929	-6 4789	-8.0737	43 4	-0.0711		
82 0	-0.7113	-0 6050	84.0	-5 6550	-5 5752	-6 0803	-5.2829	-5 7878	50.4	-0.1497		
84 0	-0 6050		87.0	1 0695	-6.5321	-5 4158	-3 2097	-4.3261	56.1	-0.0318		
87 0	-0.5518		89.0	-2.6875	-3.8067	-6.0771	-6.0131	-3.1032	62.5	0.0468		
89 0	-0 5426	-0 6605	93.0	-1.5363	-1 8882	-2.3358	-0.7370	-5.9171	69.7	0.1253		
93 0	-0.6212		96.0	-1.7602	-2.4957	-0.4811	-0 8969	-6.8126	76.9	0.1646		
96 0	-0 6212	-0 6212	100.0	-1 6323	-1.6323	-1.7283	-1 8561	-0 6410	83.4	0.2825		
100.0	-0.6212	-0 6212	96.0	0.4781	0 6381	0.5421	0.5101	-0 0654	89.4	0.2825		
96 0	-0 6605	-0 6212	84.0	0 6381	0 7979	0.7660	0.6381	-0 0015	95.4	0.0075		
84.0	-0.6212	-0.6212	73.5	0 5420	-0.6519	0 6912	0 5420	-0.0550	101.1	-0.8177		
73 0	-0.6605	-0 6212	54 0	0.3928	0 3928	0.3928	0 3928	-0 0550				
50 0	-0 6212	-0.6212	33.0	0 2435	0.2435	0.2435	0.2435	-0.0550				
35 0	-0.7407	-0.7040	24.0	0.2155	0.1123	0 1640	0.3188	-0.0941				
25 0	-0.6671	-0 7040	10 0	0 2672	0.2155	0 2155	0.3188	0 2435				
10 0	-0 6671	-0 7040	5 0	0.2672	0.2672	0 3188	0.4220	0 3928				
5 0	-0 5935	-0.6671	2.5	0 3188	0 2672	0.4220	0.4220	0.5420				
2 5	-0.6304	-0 6671										

Force Data -  
Use Run 311

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TABLE A14. (Continued)

RUN SEQ				PROPLUSIVE		WING / CANARD		PRESSURES			PAGE 273
145 3	CLAERO =	1.7577	CDAERO =	1.2786	DCLC =	0 0	DWLC =	1 8619	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4 01	2107 51	2100 62	6 89	76.93	0 0	2 66	0 0	2 071	2 071	87 0623	0.478439E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0 0	3 62	0 37	-2 50	0.03	-0 01	0 06	2 11	-1 37	2.05	0 93	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	-0 6730	-0 6374	0 0	-1.4454	-3.6920	-5 9386	-3 8916	-2.7434	38 2	-0.1393	1	-1 1646
2 5	-0 6730	-0 6374	2 5	-0.6466	-1 7949	-3 8417	-3 9915	-2 5936	43.4	-0.1393	2	0.4234
5 0	-0 6374	-0 6730	5 0	-1.1459	-1 5452	-2 1443	-3.9915	-2.5437	50 4	-0 2153	3	-1.7419
10 0	-0 6730	-0 6374	10 0	-1 0960	-1 2956	-1 7450	-3 2426	-2 5437	56 1	-0.2533	4	0 2791
15 0	-0 6730	-0 6374	15 0	-1.0459	-1.1958	-1 4953	-1.7949	-2 5437	62 5	-0.3293	5	-2.0308
25 0	-0 6374	-0 6730	24 0	-0.9960	-1.1958	-1.3955	-1.0960	-2.4439	69 7	-0.4434	6	-0.2983
35 0	-0 6374	-0 6374	33 0	-1.0201	-1.1646	-1 3089	-1 1646	-2 1751	76.9	-0 5194	7	-0 0198
50 0	-0 6374		54 0	-1.3089	-1.4533	-1 5976	-2 0308	-2 3194	83.4	-0 5574	8	-1 3807
56 0	-0 6730	-0 6374	65 0	-1 9256	-2 1916	-2 3057	-2.4197	-3 2178	89.4	-0 6334	9	-1 0404
65 0	-0 6374	-0 6730	78.5	-5.9542	-6.6004	-7 1324	-7 6645	-9 4128	95.4	-0 5574	10	0 9081
76 0	-0 6730	-0 6374	79 5*****	-136.6717	-129.7044	-124.0484	-85 5359	101.1	-0.5194	11	-0 1744	
79 0	-0.6730	-0 6374	80 5	-38.9260	-16 1208	-24.4256	-26 1228	0 1268	---	---BOTTOM---	12	-0.4837
80 5	-0.6187	-0 5930	81.3	3 1861	129.8291	-3.4210	-4 8608	-16.1730	38.2	-0.0633		
81 0	-0.5930	-0 6187	82.0	11.6959	-5 2978	6.7597	-5.9920	-9 4112	43.4	-0.0253		
82 0	-0.5930	-0 6187	84.0	-5.4007	-5 6064	-5 8378	-5.2207	-7 0718	50 4	-0.1013		
84.0	-0 6187		87 0	0.4353	-6 4804	-5 3235	-3.4210	-5 0922	56.1	0.0508		
87 0	-0 5930		89 0	-2.7107	-3 8861	-6 1129	-6.0821	-3 9788	62.5	0.1268		
89 0	-0.5574	-0 5954	93.0	-1.6282	-2 0612	-2 5251	-0.7931	-6 9481	69.7	0.2408		
93 0	-0 5954		96.0	-1.9065	-2 5869	-0 6384	-1.1333	-7 8760	76.9	0.2788		
96 0	-0 5954	-0 5954	100.0	-1.3498	-1 4736	-1 5972	-1.7829	-0 6074	83.4	0.3548		
100 0	-0 5954	-0 6334	96 0	0.5679	0 6607	0 5370	0 4751	-0.1435	89.4	0.3548		
96 0	-0 5954	-0 5954	84 0	0 6607	0 8463	0 7844	0.6298	-0 0198	95.4	0.0888		
84 0	-0 5954	-0 6334	73 5	0 7121	-0 5871	0 7121	0.5678	-0 0096	101.1	-0.7474		
73 0	-0 5954	-0.5954	54.0	0 4234	0.4234	0 4234	0.4234	-0.1540				
50 0	-0 5954	-0 6334	33.0	0 2791	0 4234	0 2791	0 2791	-0.0096				
35 0	-0 6730	-0.6374	24 0	0 2520	0 3020	0 2520	0.3519	-0.0475				
25 0	-0 6374	-0 6730	10 0	0.3519	0.3020	0 4018	0.5017	0.4234				
10 0	-0 6374	-0 6374	5.0	0 4018	0.4018	0.5017	0.5516	0.4234				
5 0	-0 6730	-0 6374	2 5	0 5017	0 4517	0 4517	0 4517	0 4234				
2 5	-0 6730	-0 6730										

Force Data -  
Use Run 311

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TABLE A14 (Concluded)

RUN SEQ 145 5  
 ALPHA 11.99  
 BETA 0 0  
 CLAERO = 1.7972  
 PTOT 2107 37  
 CL 3.64  
 PSTAT 2100 14  
 CD 0.89  
 CDAERO = 1 4909  
 Q 7.23  
 VEL 78.79  
 CM -2.41  
 O 04  
 DCLC = 0.0  
 YAW 0 0  
 H/C 2 66  
 CN -0 03  
 DWLC = 1 8464  
 CMUC 0 0  
 CMUW 1 942  
 CMUT 1 942  
 O 09  
 2 29  
 -1 33  
 BASEPR = 8 1904  
 HGT 86.9550  
 CLTR 2.08  
 RN 0.490460E+06  
 CDTR 1 21  
 PAGE 275

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**			
BP=3.375 BP=10.125			BP = 2		BP = 6		BP = 12		BP = 16		BP = 22			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	-0 6200	-0 6200	0 0	-6 8404	-9 0287	-5.0326	-4 0337	-3 2725	38.2	-0.2223	1	-1.3295		
2 5	-0 6200	-0 6200	2 5	-0 7036	-8 5055	-5 4608	-4 2714	-3 4152	43.4	-0.2585	2	0.4588		
5 0	-0 6200	-0 6200	5 0	-2 2259	-5 5085	-5 9841	-4.4618	-3.5103	50.4	-0.2585	3	-5.4563		
10 0	-0.6200	-0.6200	10.0	-1 8453	-1 6075	-6 7929	-4.9376	-3 6054	56.1	-0.4034	4	0.3212		
15 0	-0 6200	-0.5860	15.0	-1.7026	-1 6551	-5 8414	-5.7939	-3.7482	62.5	-0.5120	5	-3.8056		
25.0	-0 6200	-0 6200	24.0	-1.4647	-1.4647	-1.7026	-5.5085	-3.9861	69.7	-0.6569	6	-0.2290		
35 0	-0 6539	-0 6200	33 0	-1.3295	-1 3295	-1.0545	-3.2553	-4.4934	76.9	-0.6931	7	-0 2288		
50 0	-0 6539		54.0	-1 4671	-1 6046	-1 6046	-1.7423	-6.2817	83.4	-0.5845	8	-1.7909		
56 0	-0 6200	-0 6200	65.0	-2 0331	-2.3591	-2 2142	-1.8883	-6.2342	89.4	-0.6569	9	-1.4667		
65 0	-0 6879	-0.6539	78 5	-6 0169	-6 8136	-7.2482	-7.6104	-9.6746	95.4	-0 5845	10	0 8617		
76 0	-0.6200	-0.6200	79.5*****	-120 0598	-119.1530	-118.2960	-80 0053	101 1	-0.5120	11	-0 2878			
79 0	-0 6200	-0 6539	80.5	-57.2473	-32.7242	-46 4185	-32.6026	-0 0689	---	---	12	-0.7888		
80 5	-0 7303	-0 6814	81 3	1.9645	125 4589	-6 1198	-4.6255	-17.0948	38.2	-0 0050				
81.0	-0 6814	-0.6323	82.0	15 4387	0 8130	6.6190	-4.5765	-10 9459	43.4	0 0312				
82 0	-0 7058	-0 6568	84 0	-4 6255	-5 6299	-4 7481	-5.8993	-9 1331	50 4	-0.1137				
84 0	-0.6568		87.0	-0.0444	-6 4629	-5 6055	-4 0131	-6.7568	56 1	0 1398				
87.0	-0.6568		89 0	-2 7635	-3 9719	-6 2708	-6.5655	-5 2392	62.5	0.2847				
89 0	-0.5845	-0 6931	93 0	-1.8498	-2 1445	-2 7635	-0.8184	-7.9508	69 7	0.3934				
93 0	-0 6569		96.0	-1.9677	-2.5867	-0.9951	-1.4667	-8 0391	76.9	0.4296				
96 0	-0 6569	-0 6569	100 0	-0 5531	-1.0540	-1.2604	-1.5846	-0 6120	83.4	0.4296				
100 0	-0.6569	-0.6931	96.0	0.5964	0.7732	0 6259	0.5375	-0.0815	89 4	0.3934				
96.0	-0 6569	-0 6931	84.0	0.6849	0 8912	0 8617	0.6849	-0.1404	95.4	0.1398				
84.0	-0 6569	-0 6931	73 5	0 5963	-0 5041	0.7339	0.5963	-0.0915	101.1	-0.8018				
73 0	-0 6569	-0 6931	54 0	0.5963	0 5963	0 5963	0 3212	-0.0915						
50 0	-0 6931	-0 6569	33 0	0.4588	0.5963	0.4588	0.4588	0 0461						
35 0	-0 6200	-0 6200	24 0	0 3906	0 4381	0.4381	0.4857	0 1052						
25 0	-0 6539	-0 6200	10 0	0 5809	0 5333	0 5333	0.6284	0.3212						
10.0	-0.6539	-0.6200	5 0	0.7236	0 5809	0 4857	0.4857	0 1837						
5 0	-0 6200	-0.6200	2 5	0 6760	0 3430	0.1052	0 0576	-0 0915						
2 5	-0 6200	-0 6200												

Force Data -  
 Use Run 311

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TABLE A15. RUN 146, BW6V, DELF=45, CMU=3.0, (BN/B)=1

RUN SEQ	146 1	CLAERO = 1 9718	CDAERO = 1.3476	DCLC = 0 0	DWLC = 2.5273	BASEPR = 8.1904	PAGE 276			
ALPHA	PTOT	PSTAT	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0 05	2107 51	2102.77	4.75 63 79	0.0	2 66	0.0	2 917	2.917	86 9998	0.397616E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0 0	4 50	-0 11	-3.20	0 04	0 01	-0 02	2 28	-1 55	2.28	0.87

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	NO.	CP	
0 0	-0.8718	-0.8201	0 0	-0 3874	-1 1124	-3.1424	-4.2298	-3.5049	38 2 -0.2242	1	-1 1063	
2 5	-0 7684	-0 8201	2 5	-0 8949	-1 4748	-2 0550	-3 0699	-3 0699	43 4 -0.2242	2	0 1515	
5 0	-0 8201	-0.8201	5 0	-0 9674	-1 2574	-1.7649	-2.1274	-3.0699	50 4 -0.2794	3	-1.5256	
10 0	-0 8201	-0 8201	10 0	-0 9674	-1 1850	-1.5475	-1.6200	-2.5624	56.1 -0.2794	4	0 1515	
15 0	-0 7684	-0 7684	15 0	-0 8949	-1 1124	-1.4024	-1.5475	-1.9098	62 5 -0 3345	5	-1.5256	
25 0	-0 8201	-0 8201	24 0	-0 9674	-1 1124	-1.4024	-1 2574	-1.6924	69 7 -0 4449	6	-0.2678	
35 0	-0 7684	-0 8201	33 0	-1 1063	-1.1063	-1 3160	-1.5256	-1.5256	76 9 -0.5002	7	-0.0430	
50 0	-0.7684		54 0	-1.3160	-1 5256	-1 9448	-3.8316	-1.9448	83 4 -0 5553	8	-1.4354	
56 0	-0 7684	-0 7684	65 0	-2 1007	-2 4870	-2 5422	-2 7078	-2.9285	89.4 -0 7209	9	-1 0312	
65 0	-0 8201	-0 7684	78 5	-6 9023	-7 5095	-8 1166	-8 6133	-9.6619	95.4 -0.6105	10	0.9003	
76 0	-0.7684	-0 8201	79 5	*****-194	1544	-176 2718	-177 3167	-99.8863	101.1 -0.5553	11	-0.2226	
79 0	-0 7684	-0 7684	80 5	4 9101	-15 6237	-15 9970	-26 5994	0.1314	---BOTTOM---	12	-0.4472	
80 5	-0 8393	-0 7647	81 3	-1 4740	212 2559	-5 6180	-7.8581	-19.3944	38 2 -0.1690			
81 0	-0 7647	-0 8393	82 0	6 8888	-8 7167	9 6516	-9 1275	-9 1648	43.4 -0.1138			
82 0	-0 8393	-0 8020	84 0	-7 3355	-6 9994	-7 2607	-6.1781	-6.3275	50.4 -0.2242			
84 0	-0 7647		87 0	2 6327	-8 6049	-6 5141	-3.3781	-4 3488	56.1 -0 0034			
87 0	-0 7647		89 0	-3 0524	-4 7592	-7 9034	-7.6786	-2 8278	62 5 0.0518			
89 0	-0 6657	-0.7761	93 0	-1 6600	-2 2888	-2.6931	-0.9413	-6 7804	69.7 0.1070			
93 0	-0 7209		96 0	-2 0642	-3 2770	-0 2676	-0.6719	-8 0380	76.9 0.1622			
96 0	-0 7761	-0 7209	100 0	-2 4236	-2 4236	-2 6482	-2 7379	-0 6719	83.4 0.2725			
100 0	-0 7761	-0 7761	96 0	0 4960	0 5409	0 4061	0 4061	-0 2226	89 4 0.2173			
95 0	-0.7761	-0 7761	84 0	0 6757	0.7655	0 7655	0 6307	-0 0430	95.4 -0.0586			
84 0	-0 7761	-0 7761	73 5	0 5708	-0 8967	0 5708	0 5708	0 1515	101.1 -0.9417			
73 0	-0 7761	-0 7761	54 0	0 3611	0 3611	0 3611	0 3611	-0.0581				
50 0	-0 7761	-0 7761	33 0	0 1515	0.1515	0 1515	0 1515	-0 0581				
35 0	-0 7167	-0 7167	24 0	0 1200	0.0476	0.1200	0 3376	-0.2424				
25 0	-0 7684	-0.7684	10 0	0.1926	0.1200	0 2651	0.3376	0 1515				
10 0	-0 7684	-0 7684	5 0	0 1926	0.1926	0 2651	0 3376	0 3611				
5 0	-0 7684	-0 8718	2 5	0 3376	0 2651	0 4826	0.4826	0 3611				
2.5	-0 8201	-0 8201										

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TABLE A15. (Continued)

RUN SEQ 146 3    CLAERO = 2 0941    CDAERO = 1.5607    DCLC = 0.0    DWLC = 2.6791    BASEPR = 8.1904    PAGE 278  
 ALPHA    PTOT    PSTAT    Q    VEL    YAW    H/C    CMUC    CMUW    CMUT    HGT    RN  
 4 01    2102 51    2102 65    4.86    64 52    0.0    2.66    0.0    2 980    2.980    87.0072    0.402709E+06  
 BETA    CL    CD    CM    CROLL    CN    CY    CNTR    CMTR    CLTR    CDTR  
 0.0    4 77    0.25    -3.27    0 04    0.01    -0.02    2.60    -1.66    2.53    1.06

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
%X/C	BP=3.375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	XLOC	CP	NO.	CP
0.0	-0.7804	-0.7298	0 0	-1.6324	-4 1106	-5.9516	-4.1106	-2.9070	38.2	-0.2523	1	-1.2647	
2.5	-0.7804	-0.7298	2 5	-0 7826	-2 1280	-4.6772	-3.7567	-2.6944	43.4	-0.2523	2	0.3733	
5.0	-0 7298	-0 6794	5.0	-1.2783	-1.7032	-4.0398	-3.7567	-2.6944	50.4	-0.2523	3	-2.4932	
10 0	-0.7804	-0 7298	10.0	-1.2783	-1.4908	-1.7739	-3.8982	-2.7652	56.1	-0.3062	4	0.1686	
15 0	-0.7298	-0.7298	15.0	-1.2075	-1.4199	-1.4908	-3.5442	-2.6236	62.5	-0.3601	5	-2.2885	
25.0	-0.7804	-0 7804	24.0	-1.0659	-1 2075	-1 4199	-1.4908	-2.5529	69 7	-0.5757	6	-0.4456	
35.0	-0 7298	-0.7804	33.0	-1.0599	-1.0599	-1.4694	-1.2647	-2.2885	76.9	-0.6296	7	-0.0653	
50 0	-0.7804		54.0	-1.4694	-1 4694	-1.8789	-3.7217	-2.9027	83.4	-0.6835	8	-1.5130	
56.0	-0.7804	-0.7804	65.0	-2.1928	-2.5162	-2.6241	-2 5702	-3 9178	89.4	-0.7913	9	-1.1620	
65.0	-0.7804	-0.7298	78.5	-6.8826	-7.5294	-8.1763	-8.4997	-10.9793	95.4	-0.7374	10	0.8560	
76.0	-0 7804	-0 8309	79.5*****	-189 8630	-177.6109	-174 6214	-103.8083	101.1	-0.6296	11	-0.3285		
79 0	-0.7804	-0.7804	80 5	-10.3527	-15 6035	-18.8487	-27.0163	0.1854	---BOTTOM---	12	-0.5917		
80.5	-0.7262	-0.7992	81 3	1.4616	212.6985	-4.8101	-7.5085	-20.3440	38.2	-0.0906			
81 0	-0 6533	-0 8356	82 0	8 8637	-8 4201	9.5566	-8 6388	-10 4985	43.4	-0.0906			
82 0	-0.7992	-0 7627	84.0	-7.1439	-6 8157	-7 3262	-6 3417	-7.3990	50.4	-0.1984			
84 0	-0.7262		87.0	2 3003	-8 4565	-6.7062	-3 5339	-5.3936	56.1	0.0173			
87 0	-0 6898		89 0	-3 1362	-4 7593	-7 9179	-7.4790	-3 7063	62.5	0.0712			
89 0	-0 6835	-0 7913	93 0	-1 6883	-2 3465	-2.8728	-0.9426	-8.0932	69.7	0.1790			
93 0	-0.7374		96 0	-1.9955	-3.3117	-0.4601	-0.8549	-9.3655	76 9	0.2329			
96 0	-0 7374	-0 7374	100.0	-2.1710	-2 2587	-2.4780	-2.6535	-0 7233	83.4	0.2868			
100 0	-0 7374	-0 7913	96 0	0 5050	0 5928	0.4173	0 4173	-0.2846	89 4	0.2868			
96.0	-0.7374	-0 7913	84 0	0.6805	0.7682	0 7682	0.5928	-0.1969	95.4	-0.0366			
84 0	-0 7374	-0 7913	73.5	0.7828	-0 6504	0 7828	0.5781	-0.0362	101.1	-1.0069			
73 0	-0 7374	-0 7913	54.0	0.5781	0.5781	0.3733	0.3733	-0 2410					
50 0	-0 7913	-0.7913	33 0	0 3733	0.3733	0 3733	0.1686	-0 0362					
35 0	-0 7298	-0 7804	24 0	0.2795	0.2795	0.2795	0.3503	-0.1454					
25 0	-0 7804	-0 7804	10.0	0 4211	0.3503	0.4211	0 4919	0.3733					
10 0	-0.7804	-0.7298	5 0	0 4919	0.4211	0 4919	0.4919	0.5781					
5 0	-0 7298	-0 7804	2 5	0.6335	0.4211	0.4211	0.4211	0 3733					
2 5	-0.7804	-0 7804											

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TABLE A15 (Concluded)

RUN SEQ	PROPLULSIVE WING / CANARD PRESSURES										PAGE 280			
146 5	CLAERO = 2 6435	CDAERO = 1.9917	DCLC = 0.0	DWLC = 2 8341								BASEPR = 8 1904		
ALPHA	PTOT	PSTAT	Q VEL	YAW	H/C	CMUC	CMUW	CMUT				HGT	RN	
12.04	2107 51	2102 77	4.75 63.76	0 0	2 66	0 0	2.979	2 979				86 9569	0.398030E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR				CLTR	CDTR	
0 0	5 48	1 07	-3.53	0 05	-0 01	0 01	3.35	-1 87				3.09	1.57	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC ***	
%X/C	BP=3 375	CP	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP
0 0	-0 8201	-0 7684	0.0 -7 0572	-8 5798	-5 8248	-4 5198	-3 7223	38 2	-0.1690	1	-1.5256	
2 5	-0 8201	-0.7684	2.5 -0 8224	-8 5798	-6 1874	-4 5198	-3 7223	43 4	-0 2242	2	0.3611	
5 0	-0 8201	-0.7684	5 0 -2 5624	-4 5198	-6 3323	-4 5923	-3 7223	50 4	-0.2794	3	-6.1377	
10 0	-0 8201	-0.7684	10 0 -2 1274	-2 0550	-8 0723	-5 3174	-3 7948	56 1	-0.3897	4	0.1515	
15.0	-0 8201	-0 7167	15 0 -1 9098	-1 6924	-5 9698	-6 2598	-4 0848	62 5	-0.5553	5	-4 2508	
25.0	-0.8201	-0 7684	24 0 -1 6924	-1 5475	-1 7649	-5 8248	-4 2298	69 7	-0.6657	6	-0.4774	
35 0	-0 7167	-0.7167	33 0 -1 5256	-1 5256	-1 3160	-4 6703	-4 8799	76 9	-0.7209	7	-0.2676	
50 0	-0.7167	-0.7167	54 0 -1 7352	-1 7352	-1 9448	-3 6220	-1 1063	83 4	-0 6105	8	-1 9743	
56.0	-0.7167	-0.7684	65 0 -2 4318	-2 8182	-2 6526	-1 9903	-7 3991	89 4	-0.6657	9	-1 6151	
65 0	-0 7167	-0.7167	78 5 -7.2335	-8 1166	-8 7237	-8 6685	-11 2073	95 4	-0.6105	10	0.9003	
76 0	-0.7684	-0 7684	79 5*****	-194.9759	-186.0525	-180 1163	-103.9561	101.1	-0.5002	11	-0 3574	
79 0	-0 7167	-0 7684	80 5-14 0556	-16 1463	-25 4429	-31 6762	0 1687	---	---BOTTOM---	12	-0.8066	
80 5	-0 8393	-0 8020	81.3 3 7901	220 9559	-4 0501	-7 0740	-22 0451	38 2	0.0518			
81 0	-0 7273	-0.7647	82 0 10 8462	-7 9329	9 6888	-8 8661	-13 1969	43 4	0.0518			
82 0	-0 7647	-0 8393	84 0 -7 1861	-6 9994	-7 7835	-6 8128	-9 9862	50 4	-0.0586			
84 0	-0 7273		87 0 2 3714	-8 6421	-7 1116	-4 0874	-7 4475	56 1	0.1622			
87 0	-0.8393		89 0 -3 2321	-5 0737	-8 4422	-8 2177	-5 2982	62 5	0.2725			
89 0	-0 7209	-0 7209	93 0 -1 7497	-2 5582	-3 2770	-0 8964	-9 8347	69 7	0.4381			
93 0	-0.7209		96 0 -2 1540	-3 5464	-0 7617	-1 1658	-10 6432	76 9	0.4933			
96 0	-0 7761	-0 7761	100 0 -1 5702	-1 9743	-2 2439	-2 4685	-0 7168	83 4	0 4933			
100 0	-0 7761	-0 7761	96 0 0 5409	0 6757	0 5409	0 5409	-0 1328	89 4	0 4933			
96 0	-0 7209	-0 7209	84 0 0 6307	0 9451	0 9451	0 7206	-0 2226	95 4	0 1070			
84 0	-0 7761	-0 7761	73 5 0 5708	-0 8967	0 5708	0 5708	-0 2678	101.1	-0.9417			
73 0	-0.7209	-0 7209	54 0 0 5708	0 5708	0 3611	0 3611	-0 2678					
50 0	-0.7761	-0 7209	33 0 0 3611	0 5708	0 3611	0 3611	-0 0581					
35 0	-0 7167	-0 7167	24 0 0 4100	0 6275	0 6275	0 7001	0 1926					
25 0	-0 7684	-0 7167	10 0 0 6275	0 8450	0 7001	0 7725	0 1515					
10 0	-0 7167	-0 7167	5 0 0 6275	0 7725	0 5551	0 6275	0 1515					
5 0	-0 8718	-0 7167	2.5 0 6275	0 4100	0 3376	0 2651	-0 2678					
2 5	-0 8718	-0 7167										

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TABLE A16 RUN 147, BW6V, DELF=45, CMU=4 0, (BN/B)=1

RUN SEQ	147 1	CLAERO =	1.9880	CDAERO =	1 6250	DCLC =	0.0	DWLC =	3 5098	BASEPR =	8.1904	PAGE	281
ALPHA	PTOT	PSTAT	0	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
0 04	2107.37	2103.87	3.50	54 75	0.0	2 67	0.0	4.051	4 051	87.1901	0.342321E+06		
BETA	CL	CD	CM	CROLL	CN	CV	CNTR	CMTR	CLTR	CDTR			
0.0	5 50	-0 40	-3.88	0 06	0 02	0.00	2 47	-1 63	2.47	0.93			

***** CANARD *****			***** WING *****					**FUSELAGE**		***MISC***	
%X/C	BP=3.375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP
0.0	-0 8930	-0 8930	0 0	-0 3763	-1 3585	-4.0104	-4.4034	-4 8943	XLOC	CP	
2.5	-0 8930	-0.8930	2.5	-0 7691	-1 2602	-2 2424	-2 9299	-3 6174	38.2	-0.2723	1 -1.5573
5 0	-0.9630	-0 8930	5.0	-0 8673	-1.1620	-1.9479	-2.1442	-3.1265	43.4	-0.1975	2 0.1468
10 0	-0.8930	-0.8930	10.0	-0 8673	-1 1620	-1.6532	-1.5549	-2 1442	50.4	-0.1975	3 -1.8413
15.0	-0.8930	-0.9630	15.0	-0.8673	-1.0638	-1.4567	-1.4567	-2.4390	56.1	-0 2723	4 0 1468
25.0	-0.9630	-0 8930	24 0	-0 8673	-1 0638	-1.3585	-1.2602	-1.7513	62.5	-0.2723	5 -1.8413
35 0	-0 8930	-0 8930	33 0	-0.9892	-1 2733	-1.5573	-1.5573	-1.8413	69.7	-0.4218	6 -0.4212
50 0	-0 8930		54.0	-1 5573	-1 5573	-1.8413	-5 5336	2 9870	76.9	-0.4966	7 -0.1573
56 0	-0 8930	-0 8930	65 0	-2.2164	-2.6651	-2.6651	-2 8894	-3.0390	83.4	-0.5714	8 -1.5569
65 0	-0.8930	-0.9630	78 5	-7 6001	-8 2731	-8.8713	-9.4695	-10.8902	89.4	-0.7209	9 -1.1918
76.0	-0.9630	-0 8930	79 5	*****-248	6658-227.8775	-235.4120	-125.3004	101.1	95.4	-0.5714	10 0.8163
79 0	-0 9630	-0.9630	80.5	19 2077	-20 0425	-17 0581	-30 0580	0 1896	101.1	-0.4966	11 -0 2791
80.5	-0.8220	-0.7715	81 3	-3.7556	328.7192	-6.3857	-9.8253	-24.3417	---	---BOTTOM---	12 -0.5833
81.0	-0.9232	-0.7715	82.0	9 1929	-9 8759	14.8577	-10.9887	-10.4829	38.2	-0.0480	
82.0	-0.8726	-0 8220	84.0	-8 6621	-7.9032	-8 4089	-6.9423	-6 8918	43.4	-0.0480	
84 0	-0 8726		87.0	4.6913	-10 7358	-7.8527	-3.1994	-4.5649	50.4	-0.2723	
87.0	-0.7715		89 0	-3.1999	-5 6948	-9 6502	-9.1027	-2.2264	56.1	0.0268	
89.0	-0.6461	-0 7209	93.0	-1 6178	-2.5307	-3.1391	-1.1309	-7.3986	62.5	0.1016	
93 0	-0 7957		96 0	-2 1653	-3 9302	0.0861	-0.1573	-9.2851	69.7	0.1016	
96 0	-0.7209	-0 7957	100 0	-3 3215	-3.3825	-3 6868	-3 8083	-0 8875	76.9	0.2511	
100 0	-0 7209	-0.7209	96 0	0.4512	0 5121	0.3296	0.3296	-0.3399	83.4	0.2511	
96 0	-0 7209	-0 7957	84 0	0 5729	0 6947	0.7554	0.5121	-0.2182	89.4	0.3259	
84 0	-0.7209	-0 7209	73.5	0 4309	-0 7053	0 7149	0 4309	-0 1372	95.4	-0.1227	
73 0	-0.7209	-0.7957	54.0	0.4309	0.1468	0 4309	0.1468	-0.1372	101.1	-1 0201	
50 0	-0 7209	-0 7209	33 0	0 1468	0 1468	0 1468	0 1468	-0 1372			
35 0	-0 8930	-0 9630	24.0	0 3113	0 3113	0 2131	0 4096	-0.0816			
25 0	-0 8930	-0 8930	10 0	0 3113	0 3113	0 3113	0 4096	0.1468			
10 0	-0 8930	-0 8930	5 0	0 4096	0 3113	0 4096	0 6060	0 4309			
5 0	-0 8930	-0 8930	2 5	0 4096	0 4096	0 4096	0 5077	0.4309			
2 5	-0 8930	-0 8930									

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TABLE A16 (Continued)

RUN SEQ 147 3  
 ALPHA 4 01  
 BETA 0 0  
 CLAERO = 2 1577  
 PTOT 2107 44  
 CL 5 69  
 PSTAT 2103 82  
 CD 0 03  
 CDAERO = 1 7551  
 Q VEL 3 62 55 62  
 YAW 0 0  
 CROLL 0 06  
 DCLC = 0 0  
 H/C 2 66  
 CN 0.01  
 DWLC = 3 5317  
 CMUC 0 0  
 CY -0.00  
 CNTR 2.74  
 CMTR -1.70  
 BASEPR = 8.1904  
 HGT 87 0204  
 CLTR 0.347961E+06  
 RN 0.347961E+06  
 CDTL 1.11

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **		
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	NO.	CP	
0.0	-0.7696	-0.8374	0 0	-1.6689	-3 9523	-4 6183	-4.0474	-2.9056	38.2	-0.1637	1	-1.4809
2 5	-0 8374	-0.8374	2 5	-0.7174	-2 0495	-4 5232	-3.3814	-2.7154	43.4	-0.2362	2	0.4450
5 0	-0 8374	-0 8374	5 0	-1 2883	-1 5738	-4 0474	-3.4765	-2.5252	50 4	-0.1637	3	-2.5813
10 0	-0 8374	-0 8374	10.0	-1 1932	-1 3834	-2.9056	-3.6670	-2.5252	56.1	-0.2362	4	0.1698
15 0	-0.8374	-0.8374	15.0	-1 0980	-1 1932	-1.5738	-3.4765	-2 7154	62 5	-0.3810	5	-2.3063
25 0	-0 8374	-0 8374	24 0	-1 0028	-1.1932	-1.2883	-1.8593	-2 5252	69.7	-0.4534	6	-0.3804
35 0	-0 8374	-0 8374	33 0	-1 2058	-1 4809	-1.4809	-1.4809	-2 3063	76 9	-0.5983	7	-0.1248
50.0	-0 9053		54.0	-1 7560	-1.7560	-2.0312	-5.3327	-2 8564	83.4	-0.5983	8	-1.5395
56 0	-0 8374	-0 8374	65 0	-2 1918	-2 6988	-2.6988	-2.6988	-4.0750	89 4	-0.7432	9	-1.2447
65 0	-0 8374	-0 8374	78 5	-7 4069	-8.1312	-8 7831	-9 2177	-11.6080	95.4	-0.5983	10	0.8773
76 0	-0 8374	-0 8374	79 5	*****-242	1238	-219.3879	-229.0888	-125.0222	101.1	-0.5259	11	-0.2427
79 0	-0 8374	-0 8374	80 5	19 9565	-19 9752	-18.4071	-29.7247	0 2602	---	BOTTOM---	12	-0.5963
80 5	-0 7687	-0 8667	81.3	-4 6882	311 3733	-6 5013	-9.9799	-24 6296	38.2	-0.0913		
81 0	-0 8667	-0.7197	82 0	8 7855	-9.6860	14 0771	-11 0578	-11.6458	43.4	-0.0189		
82 0	-0 8177	-0.7687	84 0	-8 4610	-7 7751	-8.5100	-6 9422	-7.5791	50.4	-0.1637		
84 0	-0 8177		87.0	4 3267	-10 5190	-7.6770	-3 4146	-5.0313	56.1	0.1260		
87 0	-0 8177		89.0	-3.1309	-5 5478	-9.6152	-8.9667	-2.8362	62.5	0.1984		
89 0	-0.5983	-0 7432	93.0	-1.4216	-2 4826	-3 1309	-1.0090	-8.5539	69.7	0.2709		
93 0	-0.7432		96.0	-2.0699	-3 7795	-0 0658	-0.3016	-10 7352	76.9	0.3433		
96 0	-0 7432	-0 6708	100 0	-2 9543	-3 1309	-3.3668	-3 5437	-0 7731	83 4	0 4157		
100 0	-0.6708	-0 6708	96 0	0 4647	0 5826	0.3469	0.3469	-0.4195	89.4	0.2709		
96.0	-0.7432	-0 7432	84 0	0 7005	0 8184	0.8184	0 6415	-0 3016	95.4	-0.0189		
84 0	-0 6708	-0 6708	73.5	0 4450	-0 9306	0 7201	0.7201	-0.1053	101.1	-1.0329		
73 0	-0.7432	-0 7432	54 0	0 4450	0 4450	0 4450	0 4450	-0 1053				
50 0	-0 6708	-0 6708	33.0	0 1698	0 1698	0 4450	0 4450	-0.1053				
35 0	-0 7696	-0 8374	24.0	0 4243	0.5194	0.5194	0.6146	0 0437				
25 0	-0 8374	-0 7696	10 0	0 5194	0 6146	0.5194	0 7097	0.1698				
10 0	-0 9053	-0.8374	5 0	0 7097	0 6146	0 6146	0.7097	0.1698				
5 0	-0 8374	-0 8374	2 5	0.7097	0 6146	0 6146	0.5194	0 1698				
2.5	-0.8374	-0 8374										

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TABLE A16 (Concluded)

RUN SEQ	CLAERO =	2.7187	CDAERO =	2.2061	DCLC =	0.0	DWLC =	3 7018	BASEPR =	8.1904	
147 5	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
ALPHA	2107.30	2103.68	3.62	55 62	0 0	2 66	0 0	3.892	3.892	87.1712	0.348040E+06
12 00	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
BETA	6 42	1.00	-4.15	0 07	0 00	-0.01	3.54	-1.96	3.27	1.65	
0 0											

\*\*\*\*\* CANARD \*\*\*\*\*

%X/C	BP=3 375 CP	BP=10 125 CP
0.0	-0.9052	-0.9052
2 5	-0.8373	-0.9052
5 0	-0.8373	-0.9052
10 0	-0.8373	-0.8373
15 0	-0.9052	-0.9052
25 0	-0.9052	-0.8373
35 0	-0.9052	-0.9052
50 0	-0.9052	-0.9052
56 0	-0.9052	-0.9052
65 0	-0.9731	-0.9052
76 0	-0.9052	-0.8373
79 0	-0.8373	-0.8373
80 5	-0.9646	-0.9156
81 0	-0.8666	-1.1606
82 0	-0.8666	-1.0136
84 0	-0.9156	
87 0	-0.9646	
89 0	-0.7431	-0.8155
93 0	-0.8155	
96 0	-0.8880	-0.8880
100 0	-0.8155	-0.8155
96 0	-0.8880	-0.8880
84 0	-0.8155	-0.8155
73 0	-0.8880	-0.8880
50 0	-0.8880	-0.8155
35 0	-0.8373	-0.9052
25 0	-0.8373	-0.9052
10 0	-0.8373	-0.8373
5 0	-0.8373	-0.9052
2 5	-0.9052	-0.9052

\*\*\*\*\* WING \*\*\*\*\*

%X/C	BP = 2 CP	BP = 6 CP	BP = 12 CP	BP = 16 CP	BP = 22 CP
0.0	-6 9018	-9.5657	-5.7600	-4.6182	-3.8571
2 5	-0 8125	-9.5657	-6.2357	-4 9989	-3.9522
5 0	-2 5252	-3 9522	-6 5211	-4.9038	-4 0474
10 0	-2.1445	-2.1445	-8.3288	-5.2842	-4 1427
15 0	-1.9543	-1 9543	-6.5211	-6 6165	-4.3329
24 0	-1.5737	-1.7640	-1 6688	-6.1406	-4.7133
33 0	-1 4808	-1 7559	-1.4808	-3.9569	-5 0575
54 0	-1 7559	-2 0311	-2 0311	-5.0575	-6.9833
65 0	-2 5539	-2.9161	-2 7712	-2.2642	-6 7550
78 5	-7.8414	-8.7106	-9 3625	-9.5798	-11.8976
79.5	*****-245 4550	-227.0817	-232.3730	-125.1188	101.1 -0.5983
80.5	16 3801	-20 1712	-20 6121	-32.4698	0.1133
81.3	-3.7084	299 0435	-5.9620	-9.9309	-26.2465
82 0	8.4917	-10 0288	13.4402	-11.4008	-14.2425
84 0	-8 5099	-8 3631	-9 0490	-7 8240	-10 6657
87 0	4.2779	-10 7639	-8.5099	-4.2472	-7.0892
89 0	-3.3078	-5.7835	-10 0866	-9 8507	-5 0763
93 0	-1 5983	-2.7773	-3.4847	-1.1857	-11.0298
96 0	-2.1879	-4.0742	-0 2426	-0.7731	-12 4444
100 0	-2 2468	-2.7184	-3 1309	-3.4256	-0 8320
96 0	0.5237	0 6416	0.3469	0.4058	-0.3016
84 0	0 6416	0 8774	0.8184	0.5827	-0 3605
73 5	0 7202	-0 9305	0.7202	0 4451	-0 3803
54 0	0 4451	0.4451	0 4451	0 4451	-0 1052
33 0	0.4451	0 4451	0 4451	0 4451	-0.1052
24 0	0 4244	0 5195	0.5195	0.6147	0.0438
10 0	0.8049	0 7098	0 6147	0 6147	0.1699
5 0	0 6147	0 5195	0.4244	0 5195	0.1699
2.5	0 1390	0.3292	0 1390	0 0438	-0.1052

\*\*FUSELAGE\*\*

XLOC	CP	NO.	CP
38.2	-0.3085	1	-1.7559
43.4	-0.3085	2	0 4451
50.4	-0.3809	3	-6.1578
56.1	-0.4534	4	0.1699
62.5	-0.5983	5	-4.5071
69.7	-0.7431	6	-0.3803
76.9	-0 7431	7	-0 3605
83.4	-0.6707	8	-2 0698
89 4	-0.7431	9	-1.7751
95.4	-0.7431	10	0.8774
101.1	-0.5983	11	-0.4194
---	BOTTOM---	12	-0.8320
38.2	-0.0913		
43.4	-0.0188		
50.4	-0.1637		
56 1	0.1261		
62 5	0.1984		
69.7	0.3433		
76.9	0.4158		
83 4	0.4158		
89.4	0.4158		
95 4	-0.0913		
101.1	-1.1777		

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TABLE A17. RUN 148, BW6V, DELF=45, CMU=2 O, (BN/B)=1

RUN SEQ	148 1	CLAERO =	1.7538	CDAERO =	1.1491	DCLC =	0 0	DWLC =	1.7430	BASEPR =	8.1904	PAGE	286
ALPHA	0 05	PTOT	2107.16	PSTAT	2101.85	Q	5 31	VEL	67 40	YAW	0 0	H/C	2 66
BETA	0 0	CL	3 50	CD	0 14	CM	-2 42	CROLL	0 04	CN	0 01	CY	0 01
										CMUC	0.0	CMUW	2 012
										CMUT	2 012	CLTR	2.08
												CDTR	0.81

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3.375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP-----	NO.	CP	
	CP	CP		CP	CP	CP	CP	CP	XLOC			
0 0	-0 6009	-0 5548	0 0	-0.0201	-0 9917	-2.3518	-3 1938	-3.1938	38 2	-0.1672	1 -1.0440	
2 5	-0 5548	-0 5548	2.5	-0 6030	-1 1859	-1 6393	-2.2223	-3 4529	43 4	-0.1672	2 0 2671	
5 0	-0 6009	-0 6009	5 0	-0 6678	-0 9269	-1 3803	-1.5745	-2.3518	50.4	-0 1672	3 -1.2312	
10 0	-0.5548	-0 6009	10.0	-0 6678	-0 8621	-1 2508	-1 3156	-2.0279	56.1	-0.2165	4 0.2671	
15 0	-0 5548	-0 6009	15.0	-0 7326	-0 8621	-1.0564	-1.1212	-1 5745	62 5	-0.2658	5 -1.2312	
25 0	-0 5548	-0 5548	24.0	-0.6678	-0 7973	-1.0564	-0.9917	-1.5098	69 7	-0.4137	6 -0 2948	
35 0	-0 6009	-0 6009	33 0	-0 8566	-0 8566	-1 2312	-1 2312	-1.2312	76.9	-0.4630	7 0 0933	
50 0	-0 5548		54 0	-1 2312	-1 2312	-1 6058	-2 3550	-1 7932	83.4	-0.4630	8 -1.2309	
56 0	-0 5548	-0 6009	65.0	-1 7450	-2 0409	-2.1395	-2 3367	-2 4846	89 4	-0 6109	9 -0 8296	
65.0	-0.6009	-0 6009	78 5	-5.9362	-6 4293	-6 9717	-7.5634	-8 4017	95.4	-0.5616	10 0.9761	
76 0	-0.6009	-0 6009	79 5	*****	-102 3826	-79 5013	-67.9270	-77 4330	101 1	-0 4630	11 -0.1876	
79 0	-0 6009	-0 6472	80 5	-11.3916	-11 6585	-17 8282	-26 8355	0 0156	---	---BOTTOM---	12 -0 3481	
80.5	-0.5514	-0 5514	81.3	-6 2551	231 4670	-8.3230	-11 6250	-15 9944	38 2	-0.0686		
81 0	-0.6181	-0.5514	82 0	0 2824	-6.4884	3 1843	-10.0574	-9.6238	43.4	-0.0193		
82 0	-0 5848	-0 4847	84 0	-5.7881	-5 4211	-6.4218	-6.1216	-6.4218	50 4	-0.1672		
84 0	-0 5514		87 0	0 7494	-5 9882	-5.5212	-3.3199	-4 3539	56.1	0.0301		
87 0	-0.5514		89 0	-2 5150	-3 4379	-5 9259	-5 8054	-3 3176	62.5	0.0794		
89.0	-0 5123	-0.6109	93 0	-1 3112	-1 7926	-2.3947	-0.7092	-5 8455	69.7	0.1780		
93 0	-0 6109		96.0	-1 5920	-2 3947	-0.7895	-1 1507	-6.2871	76 9	0.1780		
96 0	-0 6109	-0 6109	100 0	-1 6321	-1 5519	-1 6321	-1.7926	-0 5889	83 4	0 2766		
100 0	-0 6109	-0 6109	96 0	0.5749	0 6551	0 5749	0 5347	0.0130	89.4	0 3259		
96 0	-0 6109	-0 6109	84.0	0 6952	0 8156	0 8156	0 6551	0 0532	95.4	0.0301		
84 0	-0.6109	-0 6109	73.5	0 6417	-0.6694	0.6417	0.6417	0 0798	101.1	-0.7589		
73 0	-0 6109	-0 6109	54 0	0.2671	0 2671	0.2671	0.2671	-0 1075				
50 0	-0 6109	-0 6109	33 0	0 2671	0 2671	0 2671	0 2671	-0 1075				
35.0	-0 5548	-0 6009	24 0	0 2390	0 2390	0 2390	0 4333	0 1094				
25 0	-0.5548	-0 6472	10 0	0 3038	0 2390	0 4333	0.4981	0 0798				
10 0	-0.6009	-0 6009	5 0	0 3685	0 3685	0.4981	0 6276	0 2671				
5 0	-0 6009	-0 5548	2 5	0.4333	0 4333	0 6276	0 6276	0.4544				
2 5	-0.5548	-0.6009										

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TABLE A17. (Continued)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 288				
148 3	CLAERO =	2 0914	CDAERO =	1.2509	DCLC =	0.0	DWLC =	1 7088	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4 01	2106 95	2101 63	5 31	67.38	0 0	2 66	0.0	1.901	1.901	86.9275	0.422321E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0 0	3 80	0 42	-2.53	0 04	0.01	0 01	2 41	-1 48	2.34	0.98	

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **	
	BP=3 375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XL0C	CP	NO. CP
0 0	-0 7124	-0 6662	0 0	-1 2699	-3 4724	-4.7033	-4.1851	-2 6303	38 2	-0.1367	1 -1.2503
2 5	-0 6662	-0 6662	2 5	-0 6868	-1.7881	-4.1851	-3 5372	-2.5655	43 4	-0.1367	2 0 2483
5 0	-0 6662	-0 6662	5 0	-1 2050	-1 4642	-3 5372	-3.3429	-2.3710	50 4	-0.1860	3 -2 1870
10 0	-0 7124	-0 6662	10 0	-1 1402	-1 2699	-1 4642	-3 4077	-2 5655	56.1	-0.2353	4 0.2483
15 0	-0 6662	-0 6662	15 0	-1 0107	-1.2050	-1.3347	-2.7598	-2.4358	62.5	-0.3340	5 -1.9997
25 0	-0 6662	-0 6662	24 0	-1 0755	-1 1402	-1.3347	-1.1402	-2.3063	69 7	-0.3833	6 -0.3136
35 0	-0 6662	-0.6662	33.0	-0 8756	-1 0630	-1 2503	-1 2503	-2 1870	76.9	-0.4819	7 -0.0057
50 0	-0.6662		54 0	-1.2503	-1.2503	-1.6249	-2.5616	-2.3743	83.4	-0.5313	8 -1 3303
56 0	-0 6662	-0 6662	65.0	-1.8628	-2 1587	-2.3067	-2.3067	-3.1451	89 4	-0.6299	9 -0.9288
65 0	-0 6662	-0 6662	78.5	-6.1535	-6 6466	-7.4357	-7.5837	-9.2112	95.4	-0.5806	10 0.8772
76 0	-0 6662	-0 6662	79 5*****	-104.2221	-79.7026	-68.9261	-77.4325	101.1	-0.4326	11 -0 1262	
79 0	-0.6662	-0 6662	80 5-11	5460	-11 9797	-18 2852	-27.0928	-0.0366	---BOTTOM---	12 -0 4473	
80 5	-0 7705	-0 6371	81.3	-6.4419	224.0753	-8 7102	-12 0131	-16.8170	38.2	-0.0381	
81 0	-0.7038	-0.6037	82 0	0.0301	-6.6086	2.8991	-10.3116	-10.2783	43.4	0.0113	
82 0	-0.6371	-0 7372	84.0	-5 9748	-5 6078	-6 6420	-6 4085	-7 1425	50.4	-0.1367	
84 0	-0 6705		87 0	0 5306	-6.2082	-5 8080	-3.6062	-5.0406	56.1	0.0606	
87 0	-0 7038		89 0	-2.5744	-3.4975	-6.1866	-6.1064	-4.1798	62.5	0.1592	
89 0	-0 5313	-0 6299	93 0	-1.4506	-1.8519	-2.6147	-0.7683	-7.1498	69.7	0.2085	
93 0	-0 6299		96.0	-1.6512	-2 4541	-0 9288	-1 2900	-7.5513	76.9	0.3071	
96 0	-0 5806	-0.6299	100 0	-1.4909	-1 5310	-1.6112	-1.8519	-0.5677	83.4	0.3565	
100 0	-0 5806	-0 6299	96 0	0.5963	0 6765	0 5562	0.4759	-0.1262	89.4	0.4058	
96 0	-0 6299	-0 6299	84.0	0 7167	0 8371	0 7970	0 6364	-0.0459	95.4	0.1099	
84 0	-0 5806	-0 6299	73.5	0 6230	-0.6883	0 6230	0 6230	0 0610	101.1	-0.7285	
73 0	-0 6299	-0 6299	54 0	0 4356	0 4356	0 4356	0 2483	-0.1264			
50 0	-0 5806	-0.5806	33.0	0 2483	0 2483	0 2483	0 2483	0 0610			
35 0	-0 6662	-0.6662	24.0	0 2850	0 2850	0 2202	0 4146	-0 0389			
25 0	-0 6662	-0 7124	10.0	0 4146	0 3498	0 3498	0 5441	0 2483			
10 0	-0 6662	-0 7124	5 0	0 5441	0 4146	0 4793	0 5441	0 4356			
5 0	-0 6662	-0 6199	2.5	0 5441	0 4793	0 5441	0 4793	0 4356			
2.5	-0 6662	-0.6662									

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TABLE A17. (Concluded)

RUN SEQ	148 5	CLAERO =	2.5817	CDAERO =	1.7315	DCLC =	0.0	DWLC =	1.9531	BASEPR =	8.1904	PAGE	290
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
12 O2	2107 O2	2101.70	5 31	67 38	0.0	2 67	0.0	2.053	2 053	87 1829 O	422289E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
0 0	4 53	1 10	-2.83	0.05	-0.00	-0 00	3 17	-1 72	2 93	1.47			

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
%X/C	BP=3.375 CP	BP=10.125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XLOC	CP	NO.	CP
0 0	-0.6662	-0.7125	0.0	-6.4536	-7.3606	-5.2872	-3.9266	-3.4730	38.2	-0.2847	1	-1.2505
2 5	-0.6200	-0.7125	2 5	-0.5573	-7.8791	-5.5464	-4.1209	-3.5378	43.4	-0.3340	2	0.4357
5 0	-0.6200	-0.7125	5 0	-2.2419	-6.1943	-5.4169	-4.4449	-3.6026	50.4	-0.3833	3	-5.7472
10 0	-0.6662	-0.7586	10 0	-1.8532	-1.3996	-7.3606	-5.2224	-3.6026	56.1	-0.4820	4	0.2484
15 0	-0.7125	-0.7125	15 0	-1.5939	-1.5939	-6.3240	-5.6112	-3.7970	62.5	-0.6299	5	-3.8735
25 0	-0.7125	-0.7125	24 0	-1.3996	-1.4644	-1.5939	-5.5464	-4.0561	69.7	-0.7286	6	-0.3137
35 0	-0.7125	-0.7125	33.0	-1.2505	-1.4378	-1.2505	-4.2484	-4.4357	76.9	-0.7779	7	-0.2466
50 0	-0.7125		54 0	-1.4378	-1.6252	-1.6252	-2.5620	-6.4967	83.4	-0.6793	8	-1.8925
56 0	-0.7125	-0.7586	65 0	-2.0605	-2.3564	-2.2578	-1.9124	-6.3519	89.4	-0.7286	9	-1.5713
65 0	-0.7586	-0.7586	78.5	-6.3519	-7.0918	-7.8317	-8.0783	-10.2980	95.4	-0.6793	10	0.7972
76 0	-0.7125	-0.7125	79.5*****	-106.9434	-73.7099	-74.7792	-78.9833	101.1	-0.5806	11	-0.4072	
79 0	-0.6662	-0.7125	80.5	-12.2822	-12.6492	-19.0229	-29.1330	-0.1367	---	---BOTTOM---	12	-0.8086
80 5	-0.6038	-0.6038	81.3	-6.7100	217.8481	-9.5794	-13.2164	-19.2883	38.2	-0.0874		
81 0	-0.6705	-0.7039	82 0	0.1970	-7.0769	3.0332	-11.0811	-13.1498	43.4	-0.0381		
82 0	-0.8040	-0.6038	84 0	-6.1427	-6.2762	-7.4105	-6.9101	-9.6463	50.4	-0.0874		
84 0	-0.6705		87 0	0.7643	-6.7433	-6.4764	-4.2408	-7.1103	56.1	0.0606		
87 0	-0.7039		89 0	-2.5347	-3.8595	-6.9103	-7.0709	-5.9068	62.5	0.2086		
89 0	-0.6793	-0.7286	93 0	-1.4107	-2.1333	-3.0165	-0.8487	-8.5562	69.7	0.3072		
93 0	-0.7286		96 0	-1.6114	-2.6953	-1.1298	-1.6918	-8.3555	76.9	0.3566		
96 0	-0.7779	-0.7286	100 0	-0.9289	-1.3706	-1.5713	-1.8522	-0.6882	83.4	0.3566		
100 0	-0.7286	-0.7779	96 0	0.5161	0.7169	0.5563	0.4760	-0.1663	89.4	0.3072		
96 0	-0.7286	-0.7286	84.0	0.6366	0.8373	0.7972	0.5964	-0.1262	95.4	0.0606		
84 0	-0.7286	-0.7779	73.5	0.6231	-0.6884	0.6231	0.6231	-0.1264	101.1	-0.8766		
73 0	-0.7286	-0.7286	54.0	0.4357	0.4357	0.4357	0.4357	-0.1264				
50 0	-0.7286	-0.7779	33 0	0.4357	0.4357	0.4357	0.4357	0.0610				
35 0	-0.7586	-0.7586	24.0	0.4795	0.4795	0.4795	0.6738	0.1555				
25 0	-0.7125	-0.7586	10 0	0.6091	0.6091	0.6738	0.6738	0.2484				
10 0	-0.7125	-0.7586	5.0	0.7387	0.6091	0.5443	0.5443	0.2484				
5 0	-0.6662	-0.7125	2.5	0.7387	0.4147	0.2203	0.0907	-0.1264				
2 5	-0.7586	-0.7125										

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TABLE A18. RUN 149, BW6V, DELF=45, CMU=1 O, (BN/B)=1

RUN SEQ				PROPLULSIVE		WING / CANARD		PRESSURES		PAGE 291
149 1	CLAERO =	1 5773	CDAERO =	0 8964	DCLC =	0 0	DWLC =	0 9131	BASEPR =	8.1904
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT RN
0.12	2106.52	2101 21	5 31	67.39	0 0	2.66	0 0	1 053	1.053	86 9663 0.422279E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0 0	2.49	0.37	-1.68	0 03	0.01	0.02	1.91	-1.24	1.91	0.67

***** CANARD *****			***** WING *****					**FUSELAGE**			
	BP=3 375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	**MISC ***	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO CP
0.0	-0.5274	-0 5274	0 0	-0 0389	-0.2980	-2 2415	-2.9542	-3 3429	38 2	-0.0873	1 -1.0630
2 5	-0.5274	-0 5274	2 5	-0.4924	-1 0107	-1 5936	-1 9823	-3.0189	43.4	-0 1367	2 0 2483
5.0	-0.5737	-0.5737	5.0	-0 5572	-0 7515	-1 1402	-1.3994	-1 9823	50.4	-0.1367	3 -1.2503
10.0	-0.5274	-0.5274	10 0	-0 6219	-0 7515	-1 0754	-1 0754	-1.5289	56.1	-0.1860	4 0.2483
15 0	-0 5737	-0 5274	15.0	-0 6219	-0.6867	-0 9459	-0 8811	-1.2049	62 5	-0.1860	5 -1.2503
25 0	-0.5274	-0 5737	24 0	-0.6219	-0 7515	-0 8811	-0.8811	-1 0754	69.7	-0 2846	6 -0.3136
35.0	-0 5274	-0 5274	33 0	-0 8756	-0.8756	-1 2503	-1 2503	-1 2503	76.9	-0.3340	7 0.1147
50 0	-0 5737		54.0	-1.2503	-1.2503	-1 4375	-2.3742	-1.6249	83.4	-0 3832	8 -1.0893
56 0	-0 5737	-0 5274	65.0	-1 5669	-1 7148	-1 8628	-2 0601	-2 1094	89.4	-0.4819	9 -0.6881
65 0	-0 5737	-0 5737	78 5	-5 0684	-5.4630	-6 0055	-6.4000	-6.9918	95.4	-0.4326	10 0.8773
76 0	-0 5274	-0.5737	79 5	-62 4207	-58 9515	-49 4776	-44.5064	-41.4701	101.1	-0 3832	11 -0 1261
79 0	-0 5737	-0 5274	80.5	-10 0446	-10 1781	-14 5150	-17 7510	-0.2700	---BOTTOM---		
80 5	-0 6037	-0 5369	81 3	-5 8413	233 4118	-7 0090	-9 3442	-12 2799	38 2	-0 0381	
81 0	-0 5703	-0 6037	82 0	-2 6054	-6 8087	-2 1049	-7.9430	-9.4442	43.4	-0 0381	
82 0	-0 6370	-0 6037	84 0	-4 8405	-4.4734	-5 0741	-5.0741	-6.6754	50.4	-0.1367	
84 0	-0 5703		87 0	-1 2708	-4 1399	-4 3735	-3 3726	-4 8072	56.1	0 0606	
87 0	-0 5369		89 0	-2 4139	-2.8553	-4 1798	-4 3403	-3 8186	62.5	0.1099	
89 0	-0 4326	-0.5312	93 0	-1 5710	-1 7316	-2.0928	-0 6479	-4 9423	69.7	0.1592	
93 0	-0 5312		96.0	-1 5710	-1 8118	-1 0893	-1 6915	-4.7417	76 9	0.2579	
96 0	-0 5312	-0 5312	100 0	-0 7683	-0 6479	-0 6881	-0 7683	-0 5676	83 4	0.2579	
100 0	-0 5312	-0 5312	96.0	0 5160	0.6766	0 5963	0 5562	0.0344	89 4	0.3072	
96 0	-0 5312	-0 5312	84 0	0.6364	0 7568	0 7568	0.6364	0 0745	95 4	0.1592	
84 0	-0 5312	-0 5312	73 5	0 6230	-0 6883	0 6230	0 6230	0.0610	101 1	-0 5805	
73 0	-0 5312	-0 5312	54 0	0 2483	0 2483	0 2483	0.2483	-0.1263			
50 0	-0 5312	-0 5312	33 0	0 0610	0 2483	0 0610	0 2483	-0 1263			
35 0	-0 5737	-0 5274	24 0	0 2202	0 2202	0 2202	0.4794	0.0907			
25 0	-0 4812	-0.5274	10.0	0 2202	0 2850	0 4146	0.4794	0.2483			
10 0	-0 5274	-0 5274	5.0	0 3498	0 3498	0 4146	0.6090	0 2483			
5 0	-0.5737	-0 5274	2 5	0.3498	0 3498	0.5442	0.6737	0.2483			
2 5	-0 5274	-0 5737									

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TABLE A18 (Continued)

RUN SEQ	149 3	CLAERO = 1 7507	PROPLULSIVE CDAERO = 1 0034	WING / CANARD DCLC = 0 0	PRESSURES DWLC = 0 9353	BASEPR = 8.1904
ALPHA	PTOT	PSTAT	0 VEL	YAW H/C	CMUC CMUW CMUT	HGT RN
4 04	2106 31	2100 88	5.43 68 09	0 0 2 66	0 0 1 040 1.040	87 0458 0 426984E+06
BETA	CL	CD	CM	CROLL CN	CY CNTR CMTR	CLTR CDTR
0 0	2 69	0.55	-1 74	0 03 0 01	0 02 2.15 -1.31	2 10 0.80

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
%X/C	BP=3 375 CP	BP=10.125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	NO.	CP
0 0	-0 4527	-0 4980	0 0	-0 8442	-2 4301	-4 5866	-4 0156	-2.3665	38 2	-0 0670	1 -1.2057
2 5	-0.4527	-0.4980	2 5	-0 5271	-1 3517	-2 7471	-3.0008	-2.1128	43.4	-0.1154	2 0.2616
5 0	-0 4980	-0 4527	5 0	-1 0345	-1 1613	-1 7323	-3 4448	-2 1128	50.4	-0.1154	3 -1 5725
10 0	-0 4527	-0 4527	10 0	-0 8442	-1 0979	-1 4151	-2.8105	-2 1128	56 1	-0.1154	4 0 0782
15 0	-0 4527	-0 4980	15 0	-0.7174	-0 9077	-1 2249	-1 0979	-1 9859	62.5	-0.2119	5 -1.9394
25 0	-0 4980	-0 4980	24 0	-0 7174	-0 9077	-1 0979	-0 8442	-1.9225	69 7	-0 3085	6 -0 2886
35 0	-0 4527	-0 4980	33 0	-0 8389	-1.0223	-1 2057	-1 2057	-1.9394	76 9	-0 3568	7 0 0914
50 0	-0 4980	-0 4980	54.0	-1.2057	-1 2057	-1.5725	-1.9394	-1.9394	83 4	-0.3568	8 -1.1661
56 0	-0 4527	-0 5432	65 0	-1 5640	-1.7571	-1 9020	-1 9503	-2 5298	89.4	-0 4534	9 -0 7731
65 0	-0 4980	-0 4980	78 5	-4 9924	-5 4270	-6 0065	-6 6342	-7.2137	95.4	-0.4534	10 0.8774
76 0	-0 4980	-0.4980	79 5	-61 7844	-58.2900	-49 2738	-44 5704	-41.6971	101 1	-0.3568	11 -0.1443
79 0	-0 4980	-0.4980	80 5	-9.8492	-10 0125	-14 4874	-17.6232	-0.1807	---	BOTTOM---	12 -0.4195
80 5	-0 4746	-0 4420	81 3	-5.6028	216 9450	-7 0074	-9.2939	-13 1483	38.2	0.0295	
81 0	-0 5073	-0 5073	82 0	-2 5978	-6 5174	-2 1405	-7 8241	-9 9471	43.4	0.0295	
82 0	-0 5073	-0 5073	84 0	-4 7210	-4 4270	-5.1130	-5.1782	-6 9420	50 4	-0.1154	
84 0	-0 4746	-0 4746	87 0	-1 1605	-4 1004	-4 3944	-3.3818	-5 1782	56.1	0.1261	
87 0	-0 4746	-0 4746	89 0	-2 3057	-2 8558	-4 2313	-4 5064	-4 5849	62.5	0 1744	
89 0	-0.3568	-0 4051	93 0	-1.6377	-1 7555	-2 1879	-0 6159	-5 7245	69 7	0 2710	
93 0	-0 4534	-0 4534	96 0	-1 5590	-1 7948	-1 1661	-1 7555	-5 3708	76.9	0.3192	
96 0	-0 4534	-0 4534	100 0	-0 6945	-0 5766	-0 6945	-0 7731	-0.4980	83 4	0 3675	
100 0	-0 4534	-0 4051	96 0	0 5630	0 6809	0 6023	0 5237	-0.0265	89 4	0 4158	
96 0	-0 4534	-0 4051	84 0	0 6416	0 7595	0 7202	0 6416	0 0522	95 4	0.2226	
84 0	-0 4534	-0 4051	73.5	0.6285	-0 6555	0 6285	0.6285	-0 1052	101 1	-0 4534	
73 0	-0.4534	-0 4051	54 0	0 2616	0 2616	0 4450	0 2616	-0.1052			
50 0	-0 4534	-0 4534	33 0	0 2616	0 2616	0 2616	0.2616	-0 1052			
35 0	-0 4980	-0 4980	24 0	0 4244	0 4244	0 3610	0 5512	0 1072			
25.0	-0 4980	-0 4980	10 0	0 4244	0 4878	0 5512	0 6781	0 2616			
10 0	-0 4980	-0 4527	5 0	0 5512	0 5512	0 6147	0 6781	0.2616			
5 0	-0.4980	-0.4980	2 5	0 6147	0.6147	0 6781	0 6781	0 2616			
2 5	-0 4980	-0 4980									

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TABLE A18. (Concluded)

RUN SEQ			PROPLUSIVE	WING / CANARD	PRESSURES		PAGE 295				
149 5	CLAERO =	2.3183	CDAERO =	1 4012	DCLC =	0 0	DWLC =	1.0406	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12 03	2106 24	2100.93	5.31	67.37	0 0	2 66	0 0	1.094	1.094	87 1279	0.422648E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0 0	3.36	1.06	-2.00	0.05	-0.00	-0.00	2.87	-1.52	2.67	1.25	

***** CANARD *****			***** WING *****					**FUSELAGE**		****MISC.***	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	
0 0	-0.6199	-0 5737	0 0	-5.7397	-6.5171	-4.7033	-3.6666	-3.0837	38.2	-0.1860	
2.5	-0.5737	-0 5737	2.5	-0.4924	-7.1002	-4.8975	-3.7316	-3.0189	43.4	-0.2353	
5 0	-0 5737	-0 5737	5 0	-1.9823	-5.9341	-5.3510	-4.1201	-3.2779	50.4	-0 2353	
10.0	-0 5737	-0.5737	10.0	-1.6584	-1.2049	-6.3876	-4.1201	-3.0837	56.1	-0.3833	
15 0	-0 5737	-0 5737	15.0	-1.3994	-1.3994	-5.2215	-5.0920	-3.2132	62.5	-0.4819	
25 0	-0 5737	-0.5737	24 0	-1 2699	-1.2049	-1.5936	-4.7033	-3.3429	69.7	-0.6298	
35.0	-0.6199	-0 5274	33.0	-1 0630	-1.2503	-0 8756	-3.4982	-4.0602	76.9	-0 6298	
50 0	-0 5737		54 0	-1.2503	-1 4375	-1.2503	-2.1869	-5.3715	83 4	-0.5312	
56 0	-0.5737	-0 5737	65 0	-1 8628	-2.0601	-2.1094	-1.7148	-5.3643	89.4	-0.5805	
65 0	-0.5737	-0 5737	78.5	-5.5123	-6.1534	-6 8439	-7.2384	-8.3234	95.4	-0.4819	
76 0	-0.5737	-0.5737	79.5	-71.5953	-68.2259	-56 8828	-47.0758	-41.8033	101.1	-0.4326	
79.0	-0.5737	-0 6199	80.5	-10.6786	-11.3124	-16 5167	-19.1850	-0.2701	---BOTTOM---		
80.5	-0.6370	-0.6370	81 3	-6.1415	213.0636	-8.1097	-10.6118	-14.7819	38.2	0.0113	
81 0	-0.6037	-0 6704	82.0	-2 4719	-7.0756	-2 7388	-8 8104	-11.4792	43 4	0.0113	
82.0	-0.6370	-0.5703	84.0	-5.1407	-5.1407	-5 9413	-5.9413	-8.1431	50.4	-0.0874	
84.0	-0.5703		87.0	-1.1375	-4.6737	-5 0073	-4.1066	-5 9748	56.1	0.1592	
87 0	-0 6370		89.0	-2.4941	-3.2166	-4.9826	-5.3839	-5.1833	62.5	0.3072	
89.0	-0.5312	-0.5805	93 0	-1 6111	-1 9723	-2.5743	-0 6479	-6 0663	69.7	0 4058	
93 0	-0.5805		96.0	-1 6111	-2 0928	-1 4506	-2.0527	-5 3839	76.9	0 4058	
96 0	-0 5805	-0 6298	100 0	-0 2867	-0 5275	-0.7282	-0.8486	-0 5275	83.4	0.4551	
100 0	-0 5805	-0 5805	96 0	0 5562	0 7970	0.6766	0 5562	-0 0459	89 4	0 4551	
96 0	-0 5805	-0.5805	84.0	0 7167	0 9174	0 8371	0 7167	0 0344	95 4	0.2085	
84.0	-0.5805	-0.5805	73 5	0 6229	-0 5010	0 8103	0 6229	0 0610	101.1	-0.5805	
73 0	-0 5805	-0 5805	54 0	0 4357	0.6229	0.6229	0 4357	0 0610			
50.0	-0.6298	-0.5805	33.0	0 4357	0.4357	0.6229	0.4357	0.2483			
35 0	-0 5737	-0.6199	24.0	0 5442	0.6737	0 6090	0.6737	0 2850			
25.0	-0.5737	-0.5737	10.0	0.7385	0 7385	0.6737	0.7385	0 4357			
10.0	-0.5737	-0 5737	5.0	0 8681	0 7385	0.5442	0 6090	0 4357			
5 0	-0 5737	-0.5737	2.5	0 8033	0.5442	0.3498	0.4146	0 0610			
2 5	-0.5737	-0.5274									

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TABLE A19. RUN 187, BW6V, DELF=45,CMU=0, (BN/B)=0 5

RUN SEQ	187 2	CLAERD =	0 5462	CDAERD =	0 1998	DCLC =	0 0	DWLC =	0 0	BASEPR =	8 1916	PAGE	442
ALPHA	PTOT	PSTAT	0	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
0 06	2118 12	2088 18	29 95	156 24	0 0	2 68	0 0	0 0	0 0	87 6349	0.106678E+07		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR		
0 0	0 55	0.20	-0.35	0 00	-0.00	0 01	0 54	-0.34		0 54	0 20		

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC ***	
%X/C	BP=3 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	----TOP----	NO.	CP
	CP	CP		CP	CP	CP	CP	CP	XLOC		
0.0	-0.2310	-0.2146	0 0	0 4927	0 4582	0 2743	0.2053	-0.0131	38.2	-0.0866	1 -0.2643
2 5	-0.2146	-0.2228	2 5	-0.1970	-0.3119	-0.4843	-0.6798	-0.6108	43.4	-0.0778	2 0 1012
5 0	-0.2064	-0.2310	5 0	-0.1740	-0.3234	-0.3694	-0.4958	-0.6223	50.4	-0.0691	3 -0.3308
10 0	-0.2228	-0.2392	10 0	-0.2545	-0.3234	-0.3924	-0.4843	-0.4958	56.1	-0.0691	4 0.1344
15 0	-0.2146	-0.2310	15 0	-0.2545	-0.3464	-0.3924	-0.4154	-0.4613	62.5	-0.0778	5 -0.2643
25 0	-0.2146	-0.2310	24 0	-0.2314	-0.2774	-0.3234	-0.3809	-0.3464	69.7	-0.1303	6 0.0347
35 0	-0.2228	-0.2228	33 0	-0.2311	-0.2643	-0.3308	-0.3308	-0.3308	76.9	-0.1303	7 0.5071
50 0	-0.2228		54 0	-0.2643	-0.2976	-0.3641	-0.4003	4 1223	83.4	-0.1215	8 -0.3331
56 0	-0.2310	-0.2228	65 0	-0.3316	-0.3491	-0.4891	-0.5503	-0.4891	89.4	-0.2091	9 -0.0625
65 0	-0.2228	-0.2310	78 5	-0.3316	-0.6203	-1.1541	-1.0141	-0.8041	95.4	-0.2703	10 0.5784
76 0	-0.2228	-0.2310	79.5	-0.4813	-0.5346	-1.0850	-0.8956	-0.8128	101.1	-0.2266	11 0.1725
79.0	-0.2146	-0.2310	80.5	-0.4576	-0.5287	-0.9548	-0.8660	-0.2623	---	---BOTTOM---	12 -0.0696
80 5	-0.2209	-0.2150	81.3	-0.4695	52 3943	-0.9430	-0.8838	-0.8009	38.2	-0.0516	
81 0	-0.2209	-0.2150	82 0	-0.4813	-1.0850	-0.9075	-0.8779	-0.8187	43.4	-0.0166	
82 0	-0.2090	-0.2268	84 0	-0.4813	-0.5582	-0.7832	-0.8720	-0.8128	50.4	-0.0516	
84 0	-0.2150		87 0	-0.5050	-0.5819	-0.6885	-0.9016	-0.8068	56.1	0.0097	
87 0	-0.2150		89 0	-0.5538	-0.5965	-0.7033	-0.9169	-0.8101	62.5	0.0447	
89 0	-0.2091	-0.2353	93.0	-0.5467	-0.6036	-0.7247	-0.2761	-0.8101	69.7	0.0622	
93 0	-0.2266		96 0	-0.5467	-0.6036	-0.7318	-0.8600	-0.7959	76.9	0.0884	
96 0	-0.2353	-0.2266	100 0	-0.4541	-0.5609	-0.6820	-0.8528	-0.2476	83.4	0.1585	
100 0	-0.2353	-0.2266	96 0	0 3292	0.3648	0 3292	0.3363	0 2650	89.4	0.2110	
96.0	-0.2266	-0.2266	84 0	0 6567	0 6567	0 6211	0.6140	0 5000	95.4	0.1497	
84.0	-0.2353	-0.2266	73.5	0 5332	-0.1979	0 5997	0.5997	0.4667	101.1	-0.2966	
73 0	-0.2266	-0.2266	54 0	0.2009	0 2341	0 2674	0.3006	0 2009			
50 0	-0.2353	-0.2178	33 0	0 0680	0 0680	0 1012	0.1344	0 0680			
35 0	-0.2228	-0.2228	24 0	0.0214	0 0329	0 0559	0 1249	0 0559			
25 0	-0.2228	-0.2310	10 0	0 0444	0 0329	0 0789	0 1364	0 1344			
10 0	-0.2228	-0.2310	5 0	0 0789	0 0559	0 1134	0 1479	0 2009			
5 0	-0.2146	-0.2310	2 5	0 0904	0 1249	0 1708	0 2513	0.2674			
2 5	-0.2064	-0.2228									

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TABLE A19 (Continued)

RUN SEQ 187 4  
 ALPHA 4 01  
 BETA 0 0  
 CLAE RD = 0 7208  
 PTOT 2118 26  
 CL 0 72  
 PSTAT 2088 09  
 CD 0 25  
 PROPLULSIVE CDAERD = 0.2474  
 Q 30 17  
 VEL 157 01  
 CM 0 00  
 WING / CANARD DCLC = 0 0  
 H/C 2 64  
 CN -0.00  
 PRESSURES CMUC 0.0  
 CMUW 0.0  
 CMY 0.01  
 DWLC = 0.0  
 O.0  
 CNTR 0.73  
 CMTR -0.38  
 PAGE 444  
 BASEPR = 8.1916  
 HGT 86.3496  
 RN 0.106767E+07  
 CLTR 0.72  
 CDR 0.25

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XLOC	CP	NO	CP	
0 0	-0.2275	-0.2193	0.0	-0.0715	-0.3681	-1.1894	-1.6800	-2.6156	38.2	-0.0933	1	-0.3844	
2 5	-0.2193	-0.2193	2 5	-0.2198	-0.8472	-1.0982	-1.2693	-1.4176	43.4	-0.0673	2	0.1763	
5 0	-0.2112	-0.2112	5 0	-0.5164	-0.6761	-0.8472	-0.9841	-1.1552	50.4	-0.0760	3	-0.5163	
10 0	-0.2275	-0.2112	10 0	-0.4708	-0.5734	-0.7673	-0.7673	-0.8016	56.1	-0.0760	4	0.1763	
15 0	-0.2275	-0.2193	15 0	-0.4251	-0.5050	-0.6191	-0.6647	-0.6761	62.5	-0.1020	5	-0.4503	
25 0	-0.2193	-0.2193	24 0	-0.4023	-0.4479	-0.4936	-0.5278	-0.5392	69.7	-0.1715	6	0.0114	
35 0	-0.2193	-0.2275	33 0	-0.3514	-0.4173	-0.4503	-0.4833	-0.4833	76.9	-0.1628	7	0.5203	
50 0	-0.2193		54 0	-0.3514	-0.3514	-0.4503	0.0444	3.2767	83.4	-0.1368	8	-0.2853	
56 0	-0.2193	-0.2193	65 0	-0.3626	-0.3886	-0.5623	-0.6318	-0.6057	89.4	-0.2149	9	-0.0450	
65 0	-0.2193	-0.2193	78 5	-0.3452	-0.6144	-1.1529	-1.2050	-0.9184	95.4	-0.2497	10	0.5557	
76 0	-0.2193	-0.2193	79 5	-0.4619	-0.4971	-1.0435	-1.0904	-0.8907	101.1	-0.2236	11	0.1811	
79 0	-0.2112	-0.2193	80 5	-0.4619	-0.4912	-0.9436	-1.0846	-0.2151	---BOTTOM---			12	-0.0521
80 5	-0.2210	-0.2034	81 3	-0.4560	49.8194	-0.8790	-1.0904	-0.8731	38.2	-0.0499			
81 0	-0.1975	-0.1975	82 0	-0.4619	-0.9730	-0.8143	-1.1139	-0.8731	43.4	-0.0152			
82 0	-0.2151	-0.2093	84 0	-0.4677	-0.5265	-0.7203	-1.0611	-0.8731	50.4	-0.0326			
84 0	-0.2034		87 0	-0.4971	-0.5500	-0.6675	-1.0552	-0.8966	56.1	0.0456			
87 0	-0.2093		89 0	-0.5609	-0.6033	-0.7163	-1.1545	-0.9142	62.5	0.0890			
89 0	-0.2062	-0.2149	93 0	-0.5538	-0.5962	-0.7234	-0.2711	-0.9142	69.7	0.1325			
93 0	-0.2149		96 0	-0.5609	-0.5821	-0.7376	-0.9637	-0.9072	76.9	0.1498			
96 0	-0.2149	-0.2149	100 0	-0.4549	-0.5185	-0.6740	-0.9213	-0.2358	83.4	0.1932			
100 0	-0.2149	-0.2149	96 0	0.3649	0.3861	0.3507	0.3437	0.2306	89.4	0.2367			
96 0	-0.2149	-0.2149	84 0	0.6829	0.6546	0.6829	0.5910	0.4921	95.4	0.1585			
84 0	-0.2149	-0.2149	73 5	0.5721	-0.2194	0.6051	0.5721	0.4402	101.1	-0.3018			
73 0	-0.2149	-0.2149	54 0	0.2423	0.2423	0.3082	0.3082	0.1763					
50 0	-0.2149	-0.2236	33 0	0.1434	0.1763	0.1763	0.1763	0.1104					
35 0	-0.2112	-0.2112	24 0	0.1225	0.1225	0.1909	0.2023	0.1225					
25 0	-0.2193	-0.2112	10 0	0.2137	0.2023	0.2366	0.3050	0.2753					
10 0	-0.2275	-0.2030	5 0	0.2822	0.3164	0.3392	0.4077	0.3742					
5 0	-0.2193	-0.2193	2 5	0.3734	0.4077	0.4305	0.4533	0.4402					
2 5	-0.2112	-0.2193											

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TABLE A19 (Concluded)

RUN SEQ 187 6  
 ALPHA 12 03  
 BETA 0 0  
 CLAE RD = 1 1278  
 PTOT 2118 48  
 CL 1 13  
 PSTAT 2088 64  
 CD 0 43  
 CDAERD = 0 4261  
 VEL 29 83 156 26  
 CM -0.47  
 CROLL 0 01  
 WING / CANARD DCLC = 0 0  
 YAW H/C 0 0 2 66  
 CN -0 00  
 PRESSURES DWLC = 0.0  
 CMUC 0 0  
 CMUW 0 0  
 CMUT 0 0  
 CNTR 1.19  
 CMTR -0 47  
 BASEPR = 8.1916  
 HGT 87.1149  
 RN 0.105907E+07  
 CLTR 1.12  
 CDR 0 43  
 PAGE 446

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
%X/C	BP=3.375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	XLOC	CP	NO.	CP	
0 0	-0 2483	-0 2483	0.0	-3 0244	-5 3667	-7 1204	-2 8628	-1.5361	38.2	-0.1045	1	-0 6990	
2.5	-0 2401	-0 2319	2 5	-0.1746	-2 1822	-3 9589	-2 5977	-1 5245	43 4	-0.1045	2	0 3351	
5 0	-0.2319	-0 2483	5 0	-1 2361	-1 7092	-2 4244	-2.5283	-1.5361	50 4	-0.1396	3	-1 6997	
10 0	-0 2319	-0 2483	10 0	-0 9361	-1 3169	-1 4553	-2 7244	-1 4784	56 1	-0.1747	4	0 3017	
15 0	-0 2319	-0 2401	15 0	-0 7977	-1 0054	-1 2015	-2 5629	-1 4899	62 5	-0.2450	5	-1 2661	
25 0	-0 2236	-0 2401	24 0	-0 6592	-0 7746	-0.9592	-1.5938	-1.4323	69.7	-0.3240	6	0 0349	
35 0	-0.2319	-0 2483	33 0	-0 5989	-0.6657	-0.7657	-0 9992	-1.4328	76.9	-0.2889	7	0.5090	
50 0	-0 2401		54 0	-0.4321	-0.4989	-0 5656	-0.1986	2 5700	83 4	-0.2450	8	-0 2700	
56 0	-0 2319	-0.2401	65 0	-0 4470	-0 4821	-0.6490	-0.6666	-1 2550	89.4	-0.2889	9	0 0445	
65 0	-0 2401	-0 2401	78.5	-0.3680	-0 6051	-1 0179	-1 5888	-1 3780	95.4	-0.2538	10	0.5234	
76 0	-0.2483	-0 2401	79 5	-0.4831	-0.5306	-0.9109	-1 4753	-1.3684	101 1	-0.1923	11	0 0945	
79 0	-0 2401	-0 2401	80.5	-0 5009	-0.5128	-0.8158	-1 2793	-0.2455	---BOTTOM---			12	-0.0770
80 5	-0 2573	-0 2395	81 3	-0.5128	45.3174	-0.7327	-1.2080	-1 3505	38.2	0.0185			
81 0	-0 2395	-0.2276	82.0	-0 5128	-0 9228	-0 7148	-1.1961	-1 2793	43.4	0.0448			
82 0	-0 2455	-0 2455	84.0	-0.5247	-0.5128	-0 6316	-1 3149	-1.2496	50.4	0.0185			
84 0	-0 2336		87 0	-0 5128	-0 5425	-0.6554	-1 2852	-1 1961	56.1	0.1239			
87 0	-0 2276		89 0	-0.5845	-0.6059	-0.7060	-1.3063	-1 1991	62.5	0.2029			
89 0	-0 2274	-0.2274	93 0	-0 6345	-0 5845	-0 7417	-0 2914	-1.1634	69.7	0 2556			
93 0	-0 2274		96 0	-0 6345	-0 5702	-0 7560	-0.8632	-1 1348	76 9	0 2732			
96 0	-0 2274	-0 2274	100.0	-0 5416	-0 5273	-0 6774	-0.7917	-0 2557	83.4	0.2908			
100 0	-0 2362	-0 2186	96 0	0 3661	0.4376	0.3947	0.3661	0 1732	89 4	0 2995			
96 0	-0.2274	-0 2274	84.0	0.7235	0.7878	0 6591	0.6091	0.3947	95 4	0.1854			
84 0	-0 2362	-0 2186	73.5	0 6353	-0 2320	0 6353	0 6019	0 4018	101.1	-0.3153			
73 0	-0 2274	-0.2274	54 0	0 3351	0.3684	0 3684	0 3684	0 2016					
50 0	-0 2362	-0 2186	33.0	0 3017	0 3351	0.3684	0.3684	0.1683					
35 0	-0.2401	-0 2401	24.0	0 3330	0 3907	0.3907	0 4253	0.2638					
25 0	-0 2401	-0 2483	10 0	0 5176	0.5176	0 5292	0 5407	0 4018					
10 0	-0 2401	-0 2565	5 0	0 6330	0 5984	0 5753	0 5984	0 5018					
5 0	-0 2319	-0.2483	2 5	0 6907	0 5984	0.5061	0.5176	0 4351					
2.5	-0 2401	-0.2483											

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TABLE A20. RUN 188, BW6V, DELF=45, CMU=0.5, (BN/B)=0.5

RUN·SEQ			PROPLUSIVE		WING / CANARD		PRESSURES		PAGE 447		
188 1	CLAERD =	0 4560	CDAERD =	0.3093	DCLC =	0 0	DWLC =	0.4294	BASEPR = 8.1916		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0 00	2118 76	2089.15	29 61	155.87	0 0	2 68	0.0	0 496	0 496	87.7661	0.105144E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR
0.0	0 89	0.06	-0.61	0 04	- 0 00	-0 05	0 63	-0 40		0.63	0.22

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10.125
%X/C	CP	CP
0 0	-0 3398	-0.3398
2.5	-0.3233	-0 3398
5 0	-0 3398	-0 3398
10 0	-0.3398	-0.3316
15.0	-0.3316	-0.3316
25 0	-0 3398	-0 3233
35 0	-0 3233	-0 3067
50 0	-0.3150	
56.0	-0.3150	-0 3067
65.0	-0 3067	-0 3067
76 0	-0 3233	-0 3067
79 0	-0 3067	-0 3067
80 5	-0 3439	-0 3439
81 0	-0 3439	-0.3439
82 0	-0 3379	-0.3379
84 0	-0 3559	
87.0	-0.3499	
89.0	-0.3156	-0.3156
93 0	-0 3244	
96 0	-0 3244	-0.3156
100 0	-0.3156	-0.3156
96 0	-0.3244	-0.3156
84 0	-0 3156	-0 3156
73 0	-0.3244	-0 3156
50 0	-0 3156	-0 3156
35.0	-0 2984	-0 3067
25 0	-0.3067	-0.3067
10 0	-0 3067	-0 3067
5.0	-0 3482	-0 3233
2 5	-0.3398	-0.3233

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0 0	0 4567	0 3753	0.0033	-0 3687	-0.3571
2 5	-0 3106	-0 4850	-0.7756	-0 7873	-0 9965
5.0	-0.2641	-0.4385	-0.5431	-0.6361	-0 7640
10 0	-0.3571	-0 4734	-0.5664	-0.5431	-0.6245
15 0	-0 3571	-0 4385	-0 4966	-0 5315	-0.5548
24 0	-0.3687	-0 4152	-0 4385	-0.4850	-0 4734
33 0	-0.3414	-0 4086	-0.4758	-0.5094	-0.4422
54 0	-0 5094	-0 5430	-0 6102	-0.1397	2.7509
65.0	-0 6873	-0 8112	-0 8023	-0.8377	-0.7935
78 5	-0 8112	-1.9706	-1 7316	-1 4042	-1.4661
79 5	-7.8810	-11.4611	-1 2958	-1.1820	-1.5532
80 5	-7 6415	-11 1379	-1.3137	-1.1640	-0 2960
81.3	-7.6476	46.7475	-1 3916	-1.1581	-1.4514
82 0	-7.6297	-12.2153	-1.3616	-1.1640	-1.4155
84.0	-8 1025	-11 5148	-1.3556	-1 1581	-1.3916
87.0	-8 3062	-11 0241	-1.3856	-1.1820	-1.4035
89.0	-8.4457	-9 9794	-1 4527	-1.3374	-1.4527
93 0	-7.1133	-2 6915	-1.3519	-0.3940	-1.5247
96 0	-5.3776	-0 7685	-1 2654	-1 3014	-1.5175
100.0	-2.6050	-1.0782	-1 0062	-1 2006	-0.3148
96 0	0.4054	0.4630	0.2974	0.2757	0.1389
84.0	0.6214	0.6574	0.7006	0.5710	0.4414
73 5	0 5326	-0 3078	0 5662	0.5662	0.3981
54 0	0 1964	0.2300	0 2972	0 2972	0 1628
33 0	0 0956	0.0956	0 0956	0.1628	0.0620
24.0	0 0149	0 0265	0 0963	0 1312	0 0498
10 0	0.0614	0 0730	0 1196	0.1428	0 1628
5.0	0.0614	0 0847	0 0847	0.1428	0 1964
2.5	0.1196	0.1428	0.2939	0 3288	0 3309

\*\*FUSELAGE\*\*

	-----TOP----	**MISC **
XLOC	CP	NO. CP
38.2	-0.0766	1 -0.4422
43.4	-0.0589	2 0 0956
50.4	-0.0678	3 -0.5094
56.1	-0.0678	4 0.1628
62.5	-0.0855	5 -0.4086
69 7	-0.1474	6 0 0284
76.9	-0 1740	7 0.4846
83.4	-0.1651	8 -0.4012
89.4	-0 2802	9 -0 1420
95.4	-0.3067	10 0.6358
101.1	-0.2536	11 0.2325
	---BOTTOM---	12 -0 0628
38 2	-0.0412	
43.4	-0.0147	
50.4	-0.0589	
56.1	0 0207	
62 5	0.0561	
69.7	0.0738	
76.9	0.1093	
83 4	0 1624	
89.4	0.2066	
95.4	0.0207	
101.1	-0.4749	

Force Data - Use  
Run 222

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TABLE A20 (Continued)

RUN SEQ 188 3  
 ALPHA 4.06  
 BETA 0 0  
 CLAERO = 0 5531  
 PTOT 2118 90  
 CL 1 01  
 PSTAT 2089.97  
 CD 0.12  
 CDAERO = 0 3391  
 VEL 28 93 154 13  
 YAW 0 0  
 CROLL 0.04  
 WING / CANARD DCLC = 0 0  
 H/C 2 67  
 CN -0 00  
 PRESSURES DWLC = 0 4604  
 CMUC 0 0  
 CMUW 0.512  
 CMUT 0 512  
 CNTR 0.75  
 CMTR -0 40  
 BASEPR = 8.1916  
 HGT 87.3370  
 RN 0.103796E+07  
 CLTR 0.74  
 CDTR 0.26

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **											
%X/C	BP=3 375	CP	BP=10 125	CP	BP = 2	BP = 6	BP =12	BP =16	BP = 22	XLOC	CP	NO	CP								
0 0	-0	3076	-0	3076	0.0	-0	1790	-0	7620	-1	7496	-2	2969	-3	7367	38.2	-0	0970	1	-0	5490
2 5	-0	2991	-0	2991	2.5	-0	2742	-0	9524	-1	2142	-1	7496	-1	8686	43.4	-0	0879	2	0	2078
5 0	-0	3076	-0	2906	5 0	-0	5716	-0	7858	-1	0238	-1	3451	-1	4759	50.4	-0	0879	3	-0	6867
10 0	-0	2906	-0	2991	10 0	-0	5121	-0	6668	-0	8453	-1	0238	-1	0952	56.1	-0	1061	4	0	1734
15 0	-0	2991	-0	2906	15 0	-0	4883	-0	5716	-0	7739	-0	7739	-0	9048	62.5	-0	1423	5	-0	6522
25 0	-0	2821	-0	3076	24 0	-0	4407	-0	4883	-0	6549	-0	6787	-0	6906	69.7	-0	2148	6	-0	0330
35 0	-0	3161	-0	3161	33 0	-0	4458	-0	5146	-0	6178	-0	6178	-0	6522	76.9	-0	2238	7	0	4683
50 0	-0	3076			54 0	-0	5490	-0	5834	-0	6867	0	0358	2	6502	83.4	-0	2238	8	-0	4162
56 0	-0	3161	-0	3246	65 0	-0	6949	-0	8579	-0	8398	-0	8579	-0	8579	89.4	-0	3416	9	-0	2024
65 0	-0	3076	-0	3076	78 5	-0	8126	-1	8905	-1	8995	-1	3470	-1	8633	95.4	-0	3325	10	0	6157
76 0	-0	3076	-0	3161	79 5	-7	7016	-10	9613	-1	6602	-1	0965	-2	3893	101.1	-0	2963	11	0	2177
79 0	-0	3076	-0	3246	80 5	-7	3954	-10	5752	-1	6050	-1	1026	-0	2877	---BOTTOM---			12	-0	1066
80 5	-0	2877	-0	2571	81 3	-7	4258	41	1732	-1	5805	-1	0904	-1	6786	38.2	-0	0336			
81 0	-0	2693	-0	2693	82 0	-7	5115	-10	9980	-1	5928	-1	1271	-1	6663	43.4	0	0026			
82 0	-0	2877	-0	2754	84 0	-7	9159	-10	9551	-1	6357	-1	1578	-1	4519	50.4	-0	0336			
84 0	-0	2754			87 0	-8	0017	-10	6302	-1	4641	-1	1762	-1	5070	56.1	0	0570			
87 0	-0	2816			89 0	-8	3249	-9	9392	-1	4702	-1	3228	-1	5734	62.5	0	0932			
89 0	-0	2782	-0	2782	93 0	-7	2564	-3	2836	-1	3228	-0	3720	-1	6250	69.7	0	1385			
93 0	-0	2872			96 0	-5	6051	-1	3670	-1	2122	-1	2712	-1	6176	76.9	0	1657			
96 0	-0	2782	-0	2963	100 0	-3	4972	-1	1975	-1	0280	-1	1828	-0	3277	83.4	0	2110			
100 0	-0	2872	-0	2872	96 0	0	4462	0	4757	0	2914	0	2767	0	0924	89.4	0	2291			
96 0	-0	2872	-0	2963	84 0	0	6673	0	6747	0	6894	0	5715	0	3946	95.4	0	0660			
84 0	-0	2872	-0	2872	73 5	0	5518	-0	2738	0	5862	0	5518	0	3454	101.1	-0	4050			
73 0	-0	2872	-0	2963	54 0	0	2766	0	2766	0	3110	0	2766	0	1390						
50 0	-0	2782	-0	3053	33.0	0	1734	0	2078	0	2078	0	1734	0	0702						
35 0	-0	3161	-0	3246	24 0	0	1661	0	1661	0	2137	0	2375	0	1185						
25 0	-0	3076	-0	3161	10.0	0	2613	0	2375	0	2851	0	3565	0	2766						
10 0	-0	3076	-0	3331	5 0	0	3208	0	3208	0	4040	0	4516	0	4142						
5 0	-0	2906	-0	3076	2 5	0	4159	0	4398	0	4993	0	5350	0	4486						
2 5	-0	2906	-0	2991																	

Force Data - Use  
Run 222

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TABLE A20 (Concluded)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 451				
188 5	CLAERO =	0 7127	CDAERO =	0 4814	DCLC =	0 0	DWLC =	0.4882	BASEPR =	8.1916	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11 97	2118.83	2089 56	29 27	155.13	0.0	2.66	0.0	0.513	0 513	87.1025	0.104246E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR
0 0	1.20	0.32	-0 63	0.03	-0.01	-0 01	0.98	-0 42		0 91	0.42

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	----	TOP----	NO.	CP
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP		
0.0	-0.3693	-0.3693	0.0	-3.1890	-5.8704	-9.2576	-4.1652	-2.0954	38.2	-0.1125	1	-0 8724
2 5	-0 3525	-0 3776	2.5	-0.3078	-2 5071	-3.6947	-3.8594	-2.0130	43.4	-0.1125	2	0.3856
5 0	-0.3525	-0.3609	5.0	-1 2839	-1 7308	-3.2362	-3.7067	-2.0248	50.4	-0.1483	3	-2.3345
10 0	-0.3609	-0 3776	10.0	-1 0957	-1 3309	-1.9190	-3.7067	-1.9895	56.1	-0.1931	4	0.3176
15 0	-0.3776	-0 3609	15 0	-0 9899	-1.0722	-1.4720	-3.2362	-1.9190	62.5	-0 2647	5	-1 6544
25.0	-0 3609	-0 3860	24 0	-0.8135	-0.8958	-1.1428	-1.6249	-1.8601	69 7	-0.3453	6	0.0116
35 0	-0 3609	-0.3860	33.0	-0.7024	-0 8384	-0 9404	-0 8044	-1.9264	76.9	-0.3453	7	0.3274
50 0	-0.3776		54.0	-0 7024	-0.7364	-0 8724	0.1136	2.3577	83.4	-0 3094	8	-0 8455
56 0	-0 3776	-0 3776	65.0	-0.8018	-0.9899	-1.0436	-0.9988	-1.6702	89 4	-0.3811	9	-0 4157
65.0	-0.3776	-0.3693	78.5	-0 7660	-1.8224	-2.4760	-2.1895	-2.0731	95 4	-0.3542	10	0.7499
76 0	-0 3776	-0.3776	79.5	-6 0757	-9.6730	-1.4853	-2 0970	-2.6298	101.1	-0.2826	11	-0.0368
79.0	-0 3525	-0 3776	80.5	-6.6630	-9.2248	-1.4914	-1.7275	-0.3528	----	BOTTOM----	12	-0.2991
80 5	-0.3468	-0.3407	81 3	-6.1362	36.8308	-1.8729	-1.6730	-2.1515	38.2	0.0128		
81.0	-0.3407	-0.3347	82 0	-6.0214	-10 3208	-1.7094	-1.5883	-2.0788	43.4	0.0486		
82 0	-0 3468	-0.3589	84.0	-6 2937	-9 2186	-1.6004	-1 7215	-1.9637	50.4	0 0039		
84 0	-0 3347		87.0	-6.6934	-9.3761	-1 6367	-1 6367	-1.8184	56.1	0 1382		
87.0	-0.3589		89.0	-6.8048	-8 8517	-1.7270	-1.7270	-1.7853	62.5	0.2188		
89.0	-0.3363	-0 3453	93 0	-5.9015	-4 8888	-1.5449	-0.4375	-1.7124	69.7	0.2814		
93 0	-0 3542		96 0	-5 1802	-2 6666	-1.3773	-1.5230	-1 6250	76.9	0.2904		
96 0	-0 3453	-0 3542	100.0	-5 8068	-1 3700	-1.1661	-1.1879	-0 3720	83.4	0.3083		
100.0	-0.3542	-0.3542	96.0	0.4002	0.4877	0 3201	0 3420	0.0724	89.4	0 3262		
96 0	-0.3453	-0 3453	84.0	0.7062	0 8082	0.7062	0.6188	0 3420	95.4	0.1829		
84 0	-0 3542	-0.3542	73.5	0 6236	-0.3624	0.6236	0 6236	0 3516	101.1	-0.2468		
73.0	-0 3453	-0 3453	54 0	0 3516	0.3856	0 3856	0.3516	0 1476				
50 0	-0 3721	-0 3542	33.0	0 3516	0 3516	0.3516	0.3516	0 1816				
35.0	-0.3525	-0 3776	24.0	0 3273	0.3861	0.4214	0.4214	0 2567				
25 0	-0 3609	-0 3609	10.0	0 5155	0.5155	0 5390	0 5507	0 4196				
10 0	-0 3609	-0 3693	5.0	0.6213	0 5978	0.5625	0.5978	0 4876				
5.0	-0.3525	-0.3609	2.5	0.7036	0.5743	0.4096	0 4684	0.3856				
2.5	-0 3776	-0.3693										

Force Data - Use  
Run 222

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TABLE A21. RUN 189, BW6V, DELF=45, CMU=1 0, (BN/B)M=0 5

RUN SEQ			PROPLULSIVE				WING / CANARD		PRESSURES				PAGE 452
189 1	CLAERO =	0.6210	CDAERO =	0 5816	DCLC =	0.0	DWLC =	0 9068	BASEPR =	8.1916			
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
0 09	2119.11	2105.10	14.01	107 02	0.0	2 65	0.0	1.046	1 046	86 6633	0.723852E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
0 0	1 53	0 06	-1 08	0.01	0.01	-0.01	0.98	-0.64	0.98	0.39			

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC ***	
	BP=3 375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0 0	-0.4372	-0 4197	0.0	0.4341	0 3604	-0.0818	-0 4257	-0 9660	38.2	-0.1042	1 -0.4923	
2 5	-0 4372	-0 4022	2.5	-0 4257	-0.4010	-0.7204	-1 1134	-1 3098	43.4	-0.0855	2 0 1468	
5 0	-0.4197	-0 4022	5 0	-0 3274	-0 4010	-0.5975	-0 7204	-0 8677	50.4	-0.0855	3 -0.5633	
10 0	-0.4197	-0 4197	10 0	-0 3519	-0 4747	-0.5730	-0 5975	-0 6713	56 1	-0.1042	4 0.0759	
15 0	-0 4372	-0 4197	15 0	-0 3519	-0 4010	-0.5485	-0.5485	-0.5975	62 5	-0.1229	5 -0 4923	
25 0	-0 4372	-0.4022	24 0	-0 3519	-0 4010	-0 4747	-0.4747	-0.5238	69.7	-0.1977	6 0 0048	
35 0	-0.4372	-0 4547	33 0	-0 4212	-0 4923	-0 4923	-0.4923	-0 4923	76.9	-0 2351	7 0 2991	
50 0	-0.4372		54 0	-0 4923	-0 6343	-0 7053	-0.4923	4 1235	83.4	-0.2351	8 -0.7356	
56 0	-0 4197	-0.4372	65 0	-0.7773	-1.0952	-0 9456	-0.8895	-0 7773	89.4	-0.3098	9 -0 4921	
65 0	-0.4372	-0 4547	78 5	-0 9082	-2 4414	-3 1707	-1.3382	-1 3943	95.4	-0.3847	10 0.7860	
76 0	-0.4022	-0 4372	79 5-13	5971	-19 2379	-4.3891	-1.1512	-1 7204	101.1	-0.3473	11 0.1926	
79 0	-0 4372	-0 4547	80.5-12	6866	-19.0609	-3 0611	-1 1512	-0 4555	---	---BOTTOM---	12 -0.2030	
80 5	-0 4303	-0 3923	81.3-12	9014	77 8507	-2 7449	-1 1259	-1 5560	38.2	-0.0481		
81 0	-0 4682	-0.4176	82 0-12	9014	-20.0478	-2.7069	-1.1386	-1.5054	43.4	-0.0294		
82 0	-0 4303	-0 4050	84 0-13	7363	-19.6432	-2.3528	-1 1512	-1.4421	50.4	-0.1042		
84 0	-0 4176		87 0-14	2044	-18 2011	-1 2904	-1.2271	-1.3916	56.1	0.0080		
87 0	-0 4303		89 0-14	5819	-17.5332	-1.2377	-1.2529	-1.3898	62.5	0.0454		
89 0	-0 3847	-0.3660	93 0-13	2125	-7.8564	-1.1616	-0 4769	-1.4354	69.7	0.0828		
93 0	-0 3847		96 0-10	5188	-1 7702	-1.1312	-1.2377	-1.4659	76.9	0.1015		
96 0	-0 3847	-0 3847	100 0	-6 6997	-1 5724	-0 9334	-1 1616	-0.4312	83.4	0 1576		
100 0	-0.4034	-0 3847	96 0	0 3600	0 4665	0 3143	0 2687	0 1165	89 4	0.1950		
96 0	-0.4034	-0 4034	84.0	0 6186	0 6795	0 6947	0.6186	0 4665	95.4	-0 0294		
84 0	-0 4034	-0 3847	73 5	0 5019	-0 4212	0 6439	0.5729	0 3599	101.1	-0.5529		
73 0	-0 4034	-0 4034	54 0	0.2179	0 2179	0.2889	0 2179	0 1468				
50 0	-0 3660	-0.4034	33 0	0.0759	0 0759	0 0759	0 0759	0 0759				
35 0	-0 4372	-0 4372	24 0	0 0656	0 0902	0 0902	0 1639	0 0411				
25 0	-0 4372	-0 4372	10 0	0 1147	0 1147	0.1639	0.2375	0 0759				
10 0	-0 4197	-0 4547	5 0	0 1393	0 1393	0 1884	0.2867	0 2179				
5 0	-0 4197	-0 4022	2.5	0.1884	0 1884	0 2867	0 4341	0 3599				
2 5	-0 4197	-0.4372										

Force Data - Use  
Run 223

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Table A21. (Concluded)

RUN SEQ 189 4  
 ALPHA 11.96  
 BETA 0 0  
 CLAERO = 1 0191  
 PTOT 2119 11  
 PSTAT 2104.19  
 CL 0.44  
 CDAERO = 0.7481  
 Q VEL 14.92 110.38  
 CM 0 04  
 DCLC = 0 0  
 YAW H/C 0 0 2 66  
 CROLL CN 0 01  
 DWLC = 0 9518  
 CMUC 0.0  
 CMUW 1 001  
 CMUT 1 001  
 CMTR -0 71  
 BASEPR = 8.1916  
 HGT RN 86.9015 0.747689E+06  
 CLTR CDR 1.40 0 64

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**			
%X/C	BP=3.375 CP	BP=10.125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XLOC	CP	NO.	CP	
0 0	-0.4820	-0.4985	0 0	-3.1776	-5.9928	-7.2845	-3.6622	-2.1624	38 2	-0.1429	1	-1.0025	
2.5	-0.4655	-0.4820	2 5	-0.4318	-2.5084	-4.6543	-3.4545	-2.1392	43.4	-0.1429	2	0.3984	
5 0	-0.4655	-0.4820	5 0	-1.3317	-1.9085	-3.5006	-3.7313	-2.0700	50.4	-0.1780	3	-2.5368	
10.0	-0.4655	-0.4820	10 0	-1.1701	-1.4701	-2.0931	-3.8698	-2.0931	56.1	-0.2308	4	0.3317	
15.0	-0.4655	-0.4820	15 0	-1.0317	-1.2394	-1.4701	-3.3622	-2.0470	62.5	-0.2835	5	-1.8030	
25 0	-0.4655	-0.4820	24 0	-0.8471	-1.0086	-1.2624	-1.7470	-2.0008	69.7	-0.4064	6	-0.0019	
35.0	-0.4985	-0.4985	33 0	-0.8024	-0.8691	-1.1359	-0.9358	-2.1365	76.9	-0.3713	7	0.1983	
50 0	-0.4820		54 0	-0.8024	-0.8691	-1.0692	-0.3354	3.6004	83.4	-0.3537	8	-0.9880	
56 0	-0.4985	-0.4985	65.0	-0.9861	-1.3549	-1.2847	-1.1617	-2.0751	89.4	-0.4415	9	-0.6449	
65 0	-0.4985	-0.4820	78.5	-0.7401	-2.5318	-3.0938	-2.3210	-2.2156	95.4	-0.3889	10	0.7272	
76 0	-0.4985	-0.4985	79.5-12	7008	-18.2256	-3.1836	-3.7539	-2.8390	101.1	-0.3186	11	0.0412	
79 0	-0.4985	-0.4655	80.5-11	8809	-18.3087	-2.4350	-2.6489	-0.4270	---BOTTOM---			12	-0.4305
80 5	-0.4508	-0.4389	81.3-11	7268	67.3582	-2.5063	-2.1261	-2.3994	38.2	-0.0200			
81 0	-0.4508	-0.4508	82.0-12	0.827	-18.4281	-2.8628	-1.9835	-2.1974	43.4	0.0152			
82.0	-0.4508	-0.4627	84 0-12	1068	-18.2142	-2.4112	-2.1974	-2.0310	50.4	-0.0551			
84.0	-0.4508		87.0-12	6418	-17.5607	-2.2687	-2.1024	-1.9360	56.1	0.1030			
87 0	-0.4270		89.0-12	8375	-15.9673	-2.0743	-2.0600	-1.9742	62.5	0.1908			
89 0	-0.4415	-0.4767	93.0-11	9366	-7.8060	-1.6597	-0.5735	-1.9027	69.7	0.2611			
93 0	-0.4767		96 0	-9.9645	-2.7604	-1.5311	-1.4740	-1.8170	76.9	0.2611			
96 0	-0.4767	-0.4591	100 0	-7.7631	-1.7741	-0.9736	-1.1881	-0.5020	83.4	0.2962			
100 0	-0.4767	-0.4767	96 0	0.4556	0.5271	0.3270	0.2842	-0.0160	89.4	0.2962			
96.0	-0.4767	-0.4767	84 0	0.7129	0.8416	0.7415	0.6415	0.3842	95.4	0.0854			
84 0	-0.4767	-0.4767	73.5	0.6653	-0.4021	0.6653	0.6653	0.3317	101.1	-0.4942			
73 0	-0.4767	-0.4767	54.0	0.3984	0.3984	0.3984	0.3984	0.1316					
50 0	-0.4767	-0.4942	33 0	0.3984	0.3984	0.3984	0.3984	0.1316					
35 0	-0.4985	-0.4820	24.0	0.3527	0.3758	0.3989	0.4450	0.2143					
25 0	-0.4985	-0.4820	10.0	0.5143	0.5143	0.5373	0.5835	0.3984					
10 0	-0.4985	-0.4655	5.0	0.6065	0.5835	0.5835	0.5604	0.4651					
5 0	-0.4985	-0.4985	2.5	0.6988	0.5604	0.4450	0.3989	0.3984					
2 5	-0.4820	-0.4985											

Force Data - Use  
 Run 223

A-271

TABLE A22. RUN 190, BW6V, DELF=45, CMU=2 O, (BN/B)W=0 5

RUN SEQ											PAGE 456
190 1	CLAERO =	0 3952	CDAERO =	0.8036	DCLC =	0 0	DWLC =	1.7470	BASEPR =		8.1916
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0 05	2119 25	2111.91	7 35	77.35	0 0	2 64	0 0	2 016	2 016	86 2393	0.525544E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0 0	2 14	-0 20	-1 54	0 02	0 01	-0 02	1.09	-0 70	1.09	0.43	

***** CANARD *****			***** WING *****						**FUSELAGE**		***MISC.***			
BP=3 375 BP=10.125			BP = 2		BP = 6		BP =12		BP =16		BP = 22		-----TOP----	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	-0 5630	-0 5630	0 0	0 3982	0.0702	-0 3046	-0.6327	-0 7731	38.2	-0 1697	1	-0.4828		
2 5	-0 5296	-0 5630	2 5	-0 4921	-0 5389	-0 9137	-1.1480	-1 2417	43.4	-0.1341	2	0.1945		
5 0	-0 5296	-0.5630	5.0	-0.3983	-0 4921	-0 7731	-0 8199	-0.9606	50.4	-0.1341	3	-0 6182		
10 0	-0.5296	-0 5296	10.0	-0 3983	-0 5389	-0 6795	-0.6795	-0.7731	56.1	-0.1697	4	0.1945		
15 0	-0 5296	-0.5630	15 0	-0 3983	-0 4921	-0 6327	-0.6327	-0 7263	62.5	-0.1697	5	-0 4828		
25 0	-0 5630	-0 5630	24 0	-0 4452	-0 4921	-0 5857	-0 5389	-0 6327	69.7	-0.2411	6	-0.0764		
35 0	-0.5296	-0 5630	33 0	-0 4828	-0 4828	-0 4828	-0.6182	-0.4828	76.9	-0.2768	7	0 2236		
50 0	-0.5630		54 0	-0 6182	-0.7537	-0.7537	-1 8374	6.4256	83.4	-0 2768	8	-0.8212		
56 0	-0 5296	-0 5630	65 0	-0 9544	-1 4895	-1.1328	-1 0614	-0.9188	89.4	-0.3837	9	-0 6181		
65 0	-0 5296	-0 5630	78.5	-0.3837	-3 5225	-3 5580	-1.5608	-1.8105	95.4	-0 4550	10	0.7460		
76 0	-0 5630	-0 5630	79 5-24	2.531	-38 0062	-5 0728	-1.3572	-2.1052	101 1	-0 4194	11	0 1656		
79 0	-0.5296	-0 5630	80.5-22	4.196	-37 3542	-3 7940	-1.2607	-0.5609	---BOTTOM---		12	-0 2408		
80 5	-0 4886	-0 5368	81 3-23	0.947	139 4912	-3 1425	-1.3089	-2 0087	38 2	-0 0628				
81 0	-0 5127	-0 4886	82 0-23	0.716	-38 9467	-2.9254	-1.3572	-1 7914	43.4	-0.0628				
82 0	-0 5609	-0 5852	84 0-24	8.560	-37 6917	-2 4670	-1.2848	-1 6708	50.4	-0.1341				
84 0	-0 4644		87 0-25	7.013	-35 2552	-1 4537	-1.4054	-1 5984	56.1	-0.0271				
87 0	-0.5368		89.0-26	0.430	-31.0343	-1.3727	-1 4597	-1 8661	62.5	0 0442				
89 0	-0 4907	-0 5265	93 0-23	0.825	-8.5705	-1.1985	-0 5600	-1.8951	69.7	0 0442				
93 0	-0.5265		96 0-17	2.486	-0.9083	-1.0824	-1.4307	-1.7790	76.9	0.0442				
96 0	-0 5265	-0 4907	100 0 -9	0.929	-2 8238	-0 9954	-1 3436	-0 5020	83.4	0.0799				
100 0	-0 5265	-0.5265	96 0	0.2527	0.3687	0.2236	0.2236	0.0495	89.4	0.0799				
96 0	-0.4907	-0.5265	84 0	0 5139	0 6009	0 6880	0 5719	0.4268	95.4	-0.2768				
84.0	-0 5265	-0 5265	73.5	0 6009	-0.4828	0.6009	0.6009	0.4654	101 1	-0 8118				
73 0	-0 4907	-0 5265	54 0	0 3300	0 1945	0 3300	0 3300	0 1945						
50 0	-0 4907	-0 5265	33 0	0.0591	0.0591	0 1945	0.1945	0 0591						
35 0	-0 5630	-0 5630	24.0	0 0234	0 0702	0 0234	0 1170	-0.0235						
25 0	-0 5296	-0 5630	10 0	0.1170	0 0234	0.1170	0 2108	0.1945						
10 0	-0 5630	-0 5296	5.0	0 0702	0 0702	0 2108	0.3045	0 3300						
5 0	-0 5630	-0.5296	2 5	0 1170	0 1170	0 2576	0.3513	0.3300						
2 5	-0 5630	-0.5630												

Force Data - Use  
Run 223

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TABLE A22 (Continued)

RUN SEQ	190 3	CLAERO =	0.5313	COAERO =	0 8265	DCLC =	0.0	DWLC =	1.7992	BASEPR =	8.1916	PAGE	458
ALPHA	PTDT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
4 09	2119 25	2111.79	7 46	77 93	0 0	2.69	0.0	2.000	2 000	88.0241	0 529777E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
0 0	2 33	-0 05	-1.57	0 02	0 01	-0 03	1.28	-0.74	1.25	0.50			

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.4424	-0.4753
2.5	-0.5082	-0 5082
5.0	-0 4753	-0 4753
10.0	-0.4424	-0 4753
15.0	-0.4753	-0 4753
25 0	-0.4753	-0 4753
35.0	-0.5082	-0 5082
50 0	-0 4753	
56 0	-0.4753	-0.5082
65 0	-0 4753	-0 4753
76 0	-0.4753	-0.5082
79.0	-0.4753	-0 5082
80 5	-0.5391	-0.5153
81.0	-0.5153	-0 5153
82 0	-0 5391	-0.5391
84 0	-0.5153	
87.0	-0.4679	
89 0	-0 4699	-0.4699
93.0	-0.4699	
96.0	-0.4699	-0 4699
100.0	-0.5050	-0.5050
96 0	-0.4699	-0.5050
84 0	-0 5050	-0.5050
73.0	-0 4699	-0.5050
50.0	-0 4699	-0 5050
35 0	-0 4424	-0 4753
25.0	-0.5082	-0.4753
10.0	-0.4753	-0 4753
5.0	-0.4753	-0 5082
2.5	-0 4753	-0.5082

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0.2405	-0.9788	-2.2710	-2.7787	-4 2091
2 5	-0.4251	-0 9327	-1.5327	-1.7173	-2 5479
5.0	-0 5635	-0 7020	-1.2096	-1.2558	-1 5788
10.0	-0 5174	-0.6559	-0.9327	-0.9788	-1.2096
15 0	-0 5174	-0 6559	-0.8405	-0.8405	-0 9788
24.0	-0 4713	-0.6097	-0 7481	-0.7481	-0 8405
33.0	-0.4621	-0.5955	-0.7289	-0.7289	-0 7289
54.0	-0.7289	-0 7289	-0.7289	-1.5295	6 3423
65.0	-0.9969	-1.5240	-1.1374	-1.1023	-1 0320
78 5	2.0242	-3 3857	-3.6317	-1.5942	-1.8401
79.5-24.1842	-36 5411	-5.1968	-1.4422	-2.3214	101.1 -0.3645
80.5-22 5915	-35.8520	-3.8661	-1.4184	-0.5630	---
81.3-22.6867	126.9044	-3.3195	-1.3947	-2.0125	38.2 -0.0133
82.0-22 8771	-39.0602	-3 2482	-1.3471	-1 9888	43.4 -0.0133
84 0-23.9697	-37.3254	-2.5828	-1.4659	-1 8937	50.4 -0.0836
87.0-24.9913	-34.9725	-1.6323	-1.4422	-1.8461	56.1 0.0570
89.0-26 1228	-29.0106	-1 5387	-1.5959	-1.8247	62.5 0.0921
93.0-22 5500	-6 7701	-1.3387	-0.5668	-1.8532	69.7 0.1272
96.0-17.0613	-1.6530	-1 2814	-1.5101	-1 9389	76.9 0.1623
100.0 -8 5139	-3.5398	-1.1671	-1.4530	-0 5382	83.4 0.1623
96.0 0 2908	0 3766	0.2622	0.2050	0 0335	89.4 0.1623
84 0 0 5767	0 6910	0.7768	0.5767	0 4623	95 4 -0.1538
73.5 0 4718	-0 4621	0.6052	0.6052	0.4718	101.1 -0.7510
54 0 0 3384	0 3384	0.3384	0.3384	0 0715	
33.0 0 2050	0 2050	0 2050	0.2050	0 0715	
24.0 0.2210	0.2210	0 3133	0.3594	0 1287	
10.0 0.3133	0.3133	0 3133	0.4517	0.3384	
5.0 0 3594	0.4056	0 4978	0.4978	0.3384	
2.5 0 4978	0 4517	0 5440	0.5440	0.4718	

\*\*\*FUSELAGE\*\*

----	TOP----	**MISC.**
XLOC	CP	NO. CP
38.2	-0.1186	1 -0.5955
43.4	-0.0836	2 0 3384
50.4	-0.1186	3 -0 7289
56.1	-0.1186	4 0.2050
62.5	-0.1538	5 -0.7289
69.7	-0.2592	6 -0.0619
76.9	-0.2943	7 0.1765
83.4	-0.3295	8 -0.8241
89.4	-0.3295	9 -0.6811
95.4	-0.3997	10 0.7482
101.1	-0.3645	11 0.2050
---	BOTTOM---	12 -0.2524

Force Data - Use  
Run 223

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TABLE A22. (Concluded)

RUN SEQ				PROPLULSIVE		WING / CANARD		PRESSURES		PAGE 460
190 5	CLAERO =	0 8367	CDAERO =	0 9784	DCLC =	0 0	DWLC =	1 9641	BASEPR =	8.1916
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUT	HGT	RN
12 02	2119 40	2112 16	7.23	76 73	0 0	2 68	0 0	2.065 2 065	87 5690	0 521881E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	2 80	0.34	-1.68	0 04	0 00	-0 05	1 74	-0 82	1.62	0 75

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	NO	CP	
0 0	-0 5655	-0 5994	0 0	-3 2869	-6 3797	-3 9055	-2 2403	-1 7645	38 2	-0.2001	1	-0 9306
2 5	-0 5994	-0 5655	2.5	-0 4798	-2 7636	-3 6676	-2 2879	-1 7168	43 4	-0 1638	2	0 3074
5 0	-0.5655	-0 5655	5 0	-1 3838	-2 0024	-3 5250	-2 0975	-1.7168	50.4	-0.2001	3	-1 8936
10 0	-0 5655	-0 5994	10 0	-1 1459	-1 4314	-3 0491	-2.1450	-1 6217	56.1	-0.2363	4	0 3074
15 0	-0.5315	-0 5655	15 0	-1 0984	-1 2411	-2 6208	-2.3830	-1 7645	62 5	-0 3449	5	-1 4809
25 0	-0 5655	-0 5994	24 0	-0 9080	-1 0507	-1 3838	-2 1450	-1 6693	69.7	-0 4536	6	0.0323
35 0	-0 5655	-0.5994	33 0	-0 7930	-1 0683	-0 9306	-2 1687	-1 6184	76.9	-0 4173	7	0 1110
50 0	-0 5994		54.0	-0 9306	-0 9306	-1 0683	-1.2058	6.4977	83.4	-0.4173	8	-1 0680
56 0	-0.5655	-0 5655	65 0	-1.2142	-1 9387	-1.4677	-1 1418	-1 4315	89 4	-0.4536	9	-0 8616
65 0	-0 5655	-0 6332	78 5	11 7889	-3.5323	-5 1261	-1 8663	-1.2866	95 4	-0.4173	10	0 7005
76 0	-0 5655	-0 5655	79 5	-25 0249	-37.9617	-5.8897	-2.0429	-1.3324	101.1	-0 3812	11	0 1405
79 0	-0 5315	-0.5655	80 5	-23 3841	-36 4426	-4 2970	-1 6999	-0.5484	---	BOTTOM---	12	-0.4784
80.5	-0 5239	-0.5239	81.3	-23 5058	122 7427	-3.5620	-1.5529	-1 3813	38.2	0.0173		
81.0	-0 5728	-0.5239	82 0	-23 8251	-37.7903	-2 8025	-1.5039	-1 3324	43 4	0.0535		
82.0	-0 5484	-0 5484	84 0	-24.9033	-37.1530	-2 7289	-1.2833	-1 3079	50.4	-0.0914		
84.0	-0.5728		87.0	-25 8087	-34.4341	-1 9204	-1.2833	-1 2833	56.1	0.0897		
87 0	-0 4993		89 0	-26 7693	-30 1891	-1.9817	-1 3921	-1.3038	62 5	0.1984		
89 0	-0 5622	-0 5622	93 0	-24 0877	-8 4954	-1.3627	-0 6554	-1 3627	69 7	0.2708		
93 0	-0 5622		96 0	-18 9884	-1 8932	-1 2448	-1 3038	-1 3038	76.9	0.2708		
96 0	-0.5622	-0 5622	100 0	-9 7628	-3 1606	-1 0680	-1 2448	-0.5963	83 4	0.2346		
100 0	-0 5985	-0 5260	96 0	0 4352	0 4647	0 2584	0 2289	0 0226	89.4	0.1984		
96.0	-0 5985	-0 5622	84 0	0 6710	0 8184	0 7594	0 6710	0 4352	95.4	-0.1638		
84 0	-0 5985	-0 5260	73 5	0 5825	-0 5180	0 5825	0 7201	0 4450	101 1	-0.8158		
73 0	-0 5985	-0 5622	54 0	0 3074	0 3074	0 3074	0 3074	0 1698				
50 0	-0 5622	-0 5260	33 0	0 3074	0 3074	0 3074	0 3074	0 1698				
35 0	-0 5655	-0 5655	24 0	0 3767	0 4243	0 3767	0.4718	0.1864				
25 0	-0.5994	-0.5655	10.0	0 5194	0 5194	0 5670	0 6146	0 4450				
10 0	-0 5994	-0 5994	5 0	0 6622	0 6146	0 5670	0 6622	0 4450				
5 0	-0 5655	-0 5655	2 5	0 7573	0 5194	0.5194	0 5670	0.3074				
2 5	-0.5994	-0 5315										

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Force Data - Use  
Run 223

TABLE A23. RUN 216, BC1W6V, DELF=45, CMU=0, (BN/B)W=0 5

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 491				
216 2	CLAERO =	0.5263	CDAERO =	0 2154	DCLC =	0 0	DWLC =	0.0	BASEPR =	8.1924	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.12	2146 27	2126 04	20 23	128.42	0.0	2 66	0.0	0.0	0.0	86 9964	0.865035E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0	0 53	0.22	-0.20	0 00	-0 00	0.01	0.43	-0.16	0.43	0.24	

***** CANARD *****			***** WING *****					**FUSELAGE**		****-TOP----		**MISC ***	
%X/C	BP=3 375	BP=10.125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	XLOC	CP	NO.	CP	
0 0	0.4818	0 5304	0 0	-0 0004	-0 0004	0 3058	0.2548	-0 0004	38.2	-0 1620	1	-0.2264	
2 5	-0 0883	-0 1975	2 5	-0.2047	0 2888	-0 1536	-0.4429	-0 6301	43.4	-0.1620	2	0 0688	
5.0	-0.1369	-0.2946	5 0	0 1697	0.1186	-0 1877	-0.3578	-0 5790	50.4	-0.1361	3	-0.3248	
10 0	-0 1612	-0 2946	10.0	-0 2047	-0 0004	-0 2727	-0.3408	-0.4769	56 1	-0 0973	4	0.1180	
15 0	-0 2582	-0 2946	15 0	-0 0175	-0 0515	-0 2897	-0.3238	-0.4089	62.5	-0.0714	5	-0.2756	
25 0	-0 2461	-0.2703	24 0	-0 0856	-0.1536	-0 2727	-0.2727	-0.3578	69.7	-0 0844	6	0.0196	
35 0	-0.2339	-0 2582	33 0	-0.1280	-0 1772	-0.3248	-0.3248	-0.3248	76.9	-0.0973	7	0.4764	
50 0	-0 2339		54.0	-0.2264	-0 2264	-0 3740	1.9382	-0 3740	83.4	-0.0844	8	-0 3563	
56 0	-0 2582	-0.2946	65.0	-0.3045	-0.3175	-0 5117	-0 5636	-0 4988	89 4	-0.1620	9	-0 0928	
65 0	-0 3310	-0 3674	78.5	-0.3304	-0.5765	-1.1981	-1.3406	-0.8873	95.4	-0.2139	10	0.6029	
76 0	-0 2461	-0.5373	79 5	-0.5028	-0.5115	-1.2472	-1.2034	-0.8180	101 1	-0.1879	11	0.2024	
79 0	-0.4887	-0 6828	80 5	-0 5028	-0 5290	-0.9844	-1 0720	-0.2050	---BOTTOM---			12	-0.0717
80 5	-0 4765	-0 6341	81.3	-0 5203	76 9934	-0.7567	-1.1596	-0 8531	38.2	0.0063			
81 0	-0 4415	-0 6429	82.0	-0 5028	-1 3172	-0 6692	-1.1508	-0 8180	43.4	0.0192			
82 0	-0 4765	-0 6341	84.0	-0.5028	-0.5466	-0 6166	-1.1596	-0.8531	50.4	-0.0326			
84.0	-0.2839		87.0	-0.5028	-0.5641	-0 6254	-1.0895	-0.8268	56.1	0.0192			
87 0	-0 4765		89 0	-0.5987	-0 6093	-0.6831	-1.0836	-0.8833	62 5	0.0063			
89 0	-0 4988	-0.6672	93.0	-0.5987	-0.5987	-0.7041	-0.2825	-0.9045	69.7	-0.0067			
93 0	-0.4988		96.0	-0 6093	-0.5987	-0.7253	-0 8412	-0 8728	76.9	0 0451			
96 0	-0 4988	0 6538	100 0	-0.5355	-0.5461	-0 6620	-0 7885	-0.2509	83.4	0.0969			
100.0	2.0266	-0.6542	96 0	0.1813	0.2551	0 3184	0.3816	0.2551	89.4	0.1487			
96 0	0.3171	0.3430	84.0	0.3816	0.4659	0 5924	0.5713	0.5292	95.4	0.0710			
84 0	2 0266	-0 6542	73 5	0.3640	-0 2264	0.5115	0.5608	0.5115	101.1	-0.2916			
73 0	0 3171	0 3430	54.0	0 1672	0 1672	0 2164	0.2656	0 1672					
50 0	0.6927	0.6020	33 0	0.0196	-0.2264	0 0688	0 0688	0.0688					
35 0	0.5546	0 5546	24.0	-0 0345	-0.0004	0.0166	0.0676	0.0166					
25 0	0 1664	-0.3310	10.0	-0.0856	-0 1026	-0.0004	0.0506	0.0688					
10 0	0 0087	0 0209	5 0	-0 1196	-0 2047	-0.0345	0.1186	0.1672					
5 0	-0.1126	-0 0156	2.5	-0.1366	-0.3408	-0 0345	0.1697	0.2656					
2.5	-0.2218	-0.0641											

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TABLE A23 (Continued)

RUN SEQ 216 4      CLAERO = 0 7596    CDAERO = 0.2796    DCLC = 0 0    DWLC = 0.0    PAGE 493  
 ALPHA PTOT 2146.20    PSTAT Q VEL YAW H/C CMUC CMUW CMUT    BASEPR = B 1924  
 3 98                    2125 86 20.34 128 86    0 0 2 64    0.0    0 0    0 0    HGT    RN  
 BETA    CL    CD    CM    CROLL    CN    CY    CNTR    CMTR    CLTR    CDTR  
 0 0    0 76    0 28    -0 17    0.00    -0.00    0 01    0 68    -0 13    0.67    0.30

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
%X/C	BP=3' 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	NO.	CP
0 0	0 0690	-0 8359	0 0	0 1518	0 2365	-0 3220	-1.2189	-2 1326	38.2	-0.1998	1 -0.4208
2 5	-0 7152	-1 1736	2 5	-0 1866	-0 0174	-0 7450	-0.9650	-1.9634	43.4	-0 2127	2 0 1173
5 0	-0 5825	-0 8117	5 0	0 0334	-0 1189	-0 5081	-0 7958	-1 1173	50 4	-0.1611	3 -0 5186
10 0	-0 1482	-0 6670	10 0	-0 2204	-0 2035	-0 5081	-0 6097	-0.8127	56.1	-0.1225	4 0.1173
15 0	-0 5222	-0 5945	15 0	-0.1527	-0 2374	-0 4743	-0 5251	-0 6774	62.5	-0.0967	5 -0 4697
25 0	-0 4015	-0 4257	24 0	-0 2035	-0 2712	-0 4235	-0.4404	-0 5251	69.7	-0.1354	6 -0.0295
35 0	-0 3412	-0 3774	33 0	-0.2740	-0 3230	-0 4697	-0 4697	-0 5186	76.9	-0.1483	7 0 4214
50 0	-0 3291		54.0	-0 3230	-0 3719	-0 4697	1.3893	-0.5186	83 4	-0.1225	8 -0 4906
56 0	-0.3412	-0 3654	65 0	-0.3415	-0.3801	-0 5991	-0.6635	-0.6506	89 4	-0.1998	9 -0.4068
65 0	-0 4136	-0.4619	78 5	-0.3543	-0.6377	-1.1786	-1.6937	-1.0627	95.4	-0.2384	10 0.5472
76 0	-0 2447	-0.6066	79.5	-0.5523	-0 5523	-1 1619	-1.8499	-1.0400	101.1	-0.2127	11 0.3585
79 0	-0 5463	-0 7031	80.5	-0.5348	-0 5523	-0.9180	-1.4754	-0.2300	---	---BOTTOM---	12 -0.1237
80 5	-0 4826	-0 7177	81.3	-0.5523	74 2749	-0 7177	-1.4319	-1.0312	38.2	0.0320	
81 0	-0 5174	-0 7003	82 0	-0 5261	-1 2402	-0.6829	-1.4406	-1.0661	43.4	0.0449	
82 0	-0 5261	-0 7090	84.0	-0.5523	-0.6045	-0 6568	-1.5973	-1 0661	50 4	-0.0324	
84 0	-0 3345		87.0	-0 5784	-0.5958	-0.6568	-1.3970	-1 0661	56.1	0.0320	
87 0	-0 5436		89.0	-0 6374	-0 6688	-0 7212	-1.3187	-1 1824	62 5	0 0578	
89.0	-0 5475	-0.7279	93 0	-0 6583	-0 6478	-0 7422	-0 2810	-1.1720	69.7	0.0578	
93 0	-0 5347		96.0	-0.6583	-0 6269	-0.7946	-0 8260	-1.1720	76.9	0.0964	
96 0	-0 5475	0 6631	100 0	-0 6164	-0.5640	-0.7002	-0 7841	-0.2600	83.4	0.1350	
100 0	1 6419	-0 7279	96.0	0 2327	0 3270	0 3270	0.3690	0.1698	89 4	0.1866	
96 0	0.3154	0 3154	84 0	0 4423	0 5786	0 5996	0 5891	0.4738	95 4	0.0964	
84 0	1.6419	-0 7279	73 5	0 4109	-0 2740	0 5576	0.5087	0.4109	101 1	-0 3028	
73.0	0 3154	0 3154	54 0	0.1662	0 2152	0 2152	0 2641	0.1173			
50 0	0 7661	0.6244	33.0	0 0684	-0 2740	0 1173	0 1173	0.0684			
35 0	0.6239	0 5636	24.0	0 0503	0 0503	0 1180	0.1688	0 1011			
25 0	0 1896	-0.4257	10 0	0 0165	0 0165	0 1518	0 2703	0 2152			
10.0	0.1293	0 1414	5 0	-0 0004	-0 0004	0.2026	0.3718	0.2641			
5 0	-0 0034	0 1051	2.5	-0 0681	-0 0343	0.3041	0.4395	0 4109			
2 5	0 2620	0.3947									

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TABLE A23 (Concluded)

RUN·SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 495
216 7	CLAERO =	1.2267	CDAERO =	0.4785	DCLC =	0.0	BASEPR = 8.1924
ALPHA	PTOT	PSTAT	0	VEL	YAW	H/C	HGT RN
11.99	2146.20	2125 97	20.23	128 57	0.0	2.68	87.6630 0.862648E+06
BETA	CL	CD	CM	CROLL	CN	CY	CLTR COTR
0 0	1.23	0.48	-0 08	0 00	-0 00	0.01	1.13 0 48

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC ***	
	BP=3.375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0.0	-5.6256	-1.7068	0.0	0 0286	-0 8903	-4.3448	-3.4768	-1 7411	38.2	-0.3483	1 -0.5757	
2.5	-2.5682	-1 5491	2.5	-0.3287	-0.8052	-2.3877	-3 4598	-1 6560	43.4	-0.3224	2 0 2606	
5.0	-1 7311	-1 5370	5 0	-0 4138	-0.7031	-1.2817	-3 8682	-1 6220	50 4	-0.2318	3 -1 0184	
10.0	-0.1782	-1.4521	10 0	-0.2947	-0 6861	-1.0264	-2.1155	-1 6050	56 1	-0.1799	4 0 2606	
15.0	-1 1124	-1 4278	15 0	-0 4819	-0 6520	-0.9073	-0.7031	-1.6901	62.5	-0.1929	5 -1.3628	
25 0	-0.8091	-1 4521	24.0	-0 4819	-0 6010	-0.7371	-0.6690	-1.6731	69.7	-0.2447	6 0.0639	
35 0	-0 6878	-1 3793	33.0	-0 4773	-0 5265	-0.6741	-0.6249	-1 6580	76.9	-0.2577	7 0 1342	
50 0	-0 5543		54.0	-0 4773	-0.5265	-0 5757	0 8510	-1 2152	83.4	-0.2058	8 -0.6142	
56 0	-0.5422	-0 9668	65 0	-0 4908	-0.5296	-0.6980	-0.7886	-0 9311	89.4	-0.2577	9 0 4083	
65.0	-0.5786	-0 8334	78 5	-0 4390	-0 6980	-1.0865	-1.7341	-1.2678	95.4	-0.2706	10 -0 0766	
76 0	-0 3238	-0 8576	79.5	-0.5953	-0 6303	-0 9894	-1.5849	-1.4185	101.1	-0.2058	11 -0 8461	
79 0	-0 6635	-0 9061	80.5	-0 5778	-0 5690	-0.8142	-1.3309	-0.2888	---BOTTOM---		12 -0.3718	
80 5	-0 6654	-0.9193	81.3	-0 5953	63.9675	-0 7004	-1.3747	-1.2784	38.2	0.0920		
81.0	-0 6478	-0 9193	82.0	-0 5953	-1.2083	-0 6741	-1 4536	-1 2784	43.4	0 0920		
82.0	-0 6391	-0.9281	84.0	-0.6303	-0 6303	-0 6654	-1 4711	-1 2083	50 4	-0.0116		
84 0	-0.3939		87 0	-0 6478	-0 6478	-0 7004	-1.4273	-1 2083	56.1	0.0790		
87 0	-0 6654		89.0	-0.7513	-0 7513	-0.7829	-1.2994	-1 2678	62.5	0.1179		
89 0	-0 6980	-0 9440	93 0	-0 7513	-0 7091	-0.8145	-0 3718	-1.2678	69.7	0.1697		
93 0	-0 6980		96 0	-0.7618	-0 6880	-0 8251	-0 9516	-1 2046	76.9	0.1826		
96 0	-0 6980	0 7007	100 0	-0.6669	-0.6037	-0 6459	-0.8989	-0.3402	83 4	0.2215		
100 0	1 2576	-0 8922	96 0	0 3028	0 3977	0.3766	0.3661	0.1764	89.4	0.2474		
96 0	0 3122	0 2733	84.0	0 5769	0.7139	0.6402	0.5664	0.3450	95.4	0.1179		
84 0	1 2576	-0 8922	73.5	0 5066	-0.2805	0.6050	0.5558	0 3590	101.1	-0.3613		
73 0	0.3122	0.2733	54.0	0 2606	0.3098	0.3590	0 3098	0.1622				
50.0	0.7784	0 6489	33 0	0.2114	-0.2805	0 2606	0.2606	0.1622				
35.0	0 6831	0 5861	24 0	0.1818	0 2158	0.3009	0.3349	0 2158				
25.0	0.3192	-0 6029	10.0	0 1988	0 2838	0.3860	0.4710	0.4082				
10.0	0.3192	0 2706	5.0	0 2158	0.3179	0.4540	0.5051	0.4574				
5 0	0.2585	0.2828	2.5	0.2498	0.3690	0 4370	0.4540	0.4082				
2 5	0 6103	0.5739										

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TABLE A24. RUN 217, BC1W6V, DELF=45, CMU=1 0, (BN/B)W=0 5

RUN SEQ		PROPLULSIVE WING / CANARD PRESSURES										PAGE 498			
217	1	CLAERD =	0.5873	CDAERD =	0 7072	DCLC =	0 5351	DWLC =	0 9015	BASEPR =	8.1924				
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN				
-0 03	2146.70	2138 78	7 91	80 19	0 0	2 65	0 653	1.041	1 695	86 8000	0.541223E+06				
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR					
0 0	2 02	-0 19	-0 44	-0 01	0 01	0 06	0 93	-0.10	0.93	0.46					
***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **					
BP=3'375		BP=10 125		BP = 2		BP = 6		BP =12		BP =16		BP = 22		-----TOP----	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	CP	CP	XLOC	CP	NO	CP
0.0	0 1920	-1 2969	0 0	-0 8290	-0.8290	-0.8725	0 6940	-0 5244	38.2	-0 4585	1	-0 2594			
2 5	-0 7696	-2 0416	2.5	-0 3938	0 9116	0 2153	-0 8290	-1 2641	43.4	-0.4585	2	-0 7625			
5.0	-0 6766	-1 1109	5 0	0.7375	0 7375	-0 0457	-0.6113	-1.0029	50.4	-0.2929	3	-0 6367			
10 0	-0 2113	-1.0799	10 0	-0 3503	0 3459	-0 2198	-0 6549	-0 7854	56.1	-0 0942	4	-0 1336			
15.0	-0.7386	-1 0178	15.0	0 3894	0 2153	-0 3068	-0.5678	-0 6549	62.5	-0 0611	5	-0 5110			
25 0	-0 6456	-0 8937	24 0	0.0848	-0 0457	-0 3938	-0.4809	-0 5678	69 7	-0 0942	6	-0 1336			
35 0	-0 7076	-0 8937	33.0	-0 0078	-0 2594	-0 3852	-0 6367	-0.5110	76.9	-0.1605	7	-0 2708			
50 0	-0 8937		54 0	-0 5110	-0 6367	-0 7625	0 4954	-0 6367	83 4	-0 1936	8	-0.7265			
56 0	-1.0488	-1 2350	65 0	-0 8890	-1 2865	-0 9884	-0.9884	-0 8559	89.4	-0.2598	9	-0.4300			
65 0	-1 4831	-1 8863	78 5	-0 3923	-4 6311	-3 5052	-1 4189	-1.4520	95 4	-0 3592	10	0.8368			
76 0	-0 4284	-4 4612	79 5*****	-115 8499	-5.2940	-1 1288	-1.6438	101 1	-0.2929	11	0.1630				
79 0	-6 6637	-7 8426	80.5-57	2916	-38 1224	-3 4130	-1.1064	-0 3675	---	---BOTTOM---	12	-0.1604			
80 5	-13 3559	-28 4259	81 3-20	1414	170 9370	-2 8531	-1 1288	-1.4199	38.2	0.1375					
81 0	-7 8916	-15 8638	82 0	-6.8167	-1 1064	-2 7635	-1.0841	-1 4423	43 4	0.1707					
82 0	-5 6971	-9 5711	84 0	-1 4647	-5 8538	-2 5844	-1.1512	-1 4199	50.4	0 0051					
84 0	-1.2408		87 0	-0 5690	-4 5998	-1 5094	-1.1288	-1 3975	56.1	-0.1936					
87 0	-1 5543		89 0	-2 0742	-2 6133	-1 2656	-1 3734	-1 4812	62 5	-0.5910					
89 0	-2 1474	-3 4390	93 0	-0.8074	-1 0769	-1 1038	-0.4570	-1 5620	69.7	-0 6903					
93 0	-1 0877		96 0	-1 0230	-1.0230	-1 0499	-1 2387	-1 6160	76.9	-0.5910					
96 0	-0 0611	0 7005	100 0	-1 2387	-1 5620	-0 9152	-1 2117	-0 4030	83.4	-0.4254					
100.0	1.2304	-1 0546	96.0	0 0552	0 0282	0 1899	0.4056	0.2169	89.4	-0.1936					
96 0	0 5680	0 5018	84 0	0 0552	-0.0526	0 4056	0.8368	0 5943	95.4	-0.1274					
84 0	1 2304	-1 0546	73.5	-0 1336	-0 3852	0 1180	0.4954	0 4954	101.1	-0.4585					
73 0	0.5680	0 5018	54 0	-0 5110	-0 5110	-0 3852	-0 1336	-0 0078							
50 0	0 7668	0 6674	33 0	-0 7625	-0 3852	-0 6367	-0 3852	-0 1336							
35 0	0 6884	0 5953	24 0	-0 6984	-0 7854	-0 6984	-0.3938	-0 2198							
25 0	0 2541	-1 4211	10.0	-0 6113	-0 6984	-0 7854	-0.3068	-0.0078							
10 0	0 2230	0 1920	5 0	-0 6549	-0 6984	-0 8290	-0 3068	0 1180							
5 0	0 0369	0.1300	2 5	-0 6984	-0 7854	-0 9159	-0 2198	0 2438							
2 5	0 0990	0 4091													

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TABLE A24. (Continued)

RUN SEQ	217 4	CLAERO =	0 6140	CDAERO =	0 6680	DCLC =	0 5436	DWLC =	0.9192	BASEPR =	8.1924
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4 09	2146.84	2138 93	7 91	80 16	0 0	2 67	0.634	1 022	1.655	87.4453	0.541853E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR
0 0	2.08	-0.10	-0.25	0.02	0.00	0.03	0.97	0.09		0.94	0.47

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP-----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	-1.2969	-3 7783	0 0	-0 8723	-0 6983	-0 9158	-0.6548	-2.0470	38 2	-0.4915	1	-0.3851
2 5	-1 5759	-3.2508	2 5	-0 3067	0 7810	-0 9158	-0.7418	-1.9599	43.4	-0.4584	2	-0.8883
5.0	-1 1418	-3.2508	5.0	0 6939	0 6504	-0.9158	-0.6548	-1 2639	50.4	-0.2598	3	-0 8883
10.0	-0.2112	-2.4445	10 0	-0.3502	0.3459	-0.8288	-0.6983	-0 9158	56 1	-0.0942	4	-0.2593
15 0	-0.9556	-1 5139	15.0	0.3024	0.1718	-0.7418	-0.6983	-0.8288	62.5	-0.0611	5	-0.6366
25 0	-0.8936	-1.0486	24.0	-0 0022	-0.1327	-0.7418	-0.6983	-0 7852	69.7	-0.0611	6	-0.2593
35.0	-0 8936	-1.0797	33.0	-0.1335	-0 2593	-0.8883	-0 8883	-0.6366	76.9	-0.1273	7	0 3247
50 0	-1 0486		54.0	-0.5109	-0.5109	-0 8883	0.9985	-0.6366	83.4	-0.1604	8	-0.6455
56 0	-1.1418	-1.3899	65.0	-0.8558	-1.2531	-0.9882	-1.2200	-0 8558	89.4	-0.2598	9	-0.4838
65.0	-1.5759	-2.0412	78 5	-0 1604	-4.6636	-3 4385	-1.8159	-1.4518	95 4	-0 3259	10	0.7290
76.0	-0.3663	-4.5846	79 5-99	7144-115 4320	-5.9202	-1.3973	-1.5764	101.1	-0.2266	11	0 2978	
79.0	-6.8490	-8 0587	80.5-58	7846 -41.6552	-4 0617	-1.2406	-0.3449	---BOTTOM---			12	-0.1604
80.5	-13.4882	-28.6687	81 3-21	0108 142 4148	-3.2558	-1.2406	-1.4421	38.2	0.1707			
81.0	-8.1816	-15 9288	82.0	-6 8383	-0 7256	-2.9871	-1.2630	-1.3750	43.4	0 2369		
82 0	-5 9426	-9 8610	84.0	-1 4421	-5.9650	-2.4049	-1.3077	-1.3750	50.4	0.0714		
84 0	-1.3301		87.0	-0.8152	-4.8902	-1 9571	-1.3973	-1.3750	56.1	-0.1273		
87.0	-1.6659		89.0	-2 2086	-2.7746	-1.6427	-1.5079	-1.4271	62.5	-0.4915		
89 0	-2.2133	-3.7033	93.0	-0 9151	-1.1576	-1.1036	-0.4029	-1.4810	69.7	-0.6571		
93 0	-1 1207		96 0	-1.2116	-1 0767	-0.8881	-1.2385	-1.4810	76.9	-0.6240		
96 0	-0 0611	0 7998	100.0	-1 0767	-1.1307	-0.5647	-0 9689	-0 3760	83.4	-0.4253		
100 0	1 7269	-1 0876	96.0	-0 0795	-0.1065	0 1630	0.4325	0 2708	89.4	-0.1604		
96.0	0 6011	0.4687	84 0	0 0283	-0.1065	0 3516	0.8098	0.7020	95.4	-0.1604		
84 0	1 7269	-1.0876	73.5	-0.1335	-0 3851	-0.0077	0.4954	0.4954	101.1	-0.5246		
73 0	0 6011	0 4687	54.0	-0.5109	-0 6366	-0.5109	-0.2593	-0.0077				
50 0	0 8329	0 7005	33 0	-0.7624	-0 2593	-0 8883	-0.6366	-0.1335				
35 0	0 7193	0 5953	24.0	-0 6548	-0 6983	-0.7418	-0 5677	-0.1327				
25 0	0 2230	-1 5139	10 0	-0.6112	-0.7418	-0 7852	-0.5677	-0 0077				
10 0	0.3161	0.2230	5.0	-0.5677	-0.6983	-0 7418	-0.5677	0 1180				
5 0	0 1920	0.2230	2.5	-0 6548	-0.6548	-0 7418	-0.5243	0 3696				
2 5	0 4402	0.5332										

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TABLE A24 (Concluded)

RUN SEQ				PROPLULSIVE		WING / CANARD	PRESSURES			PAGE 502
217 6	CLAERO =	1 2688	CDAERO =	1 0045	DCLC =	0.6058	DWLC =	1.0191	BASEPR =	8 1924
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	HGT	RN
11.98	2146.70	2139 12	7.57	78 41	0.0	2 66	0 658	1 072	1 730	87 1714 0.530367E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0 0	2 89	0 42	-0 23	0.02	-0 01	0.02	1 77	0.13	1.63	0.85

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
%X/C	BP=3.375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP
0 0	-7 5192	-3 4361	0 0	-0 6782	-0 4965	-1 5419	-1 5873	-4 9055	38 2	-0.6916	1 -0.9677
2 5	-8 0052	-3.6953	2.5 0	-0 4965	0 3672	-1.3601	-1 3147	-3 6781	43 4	-0.6916	2 -0.5734
5 0	-3.4361	-3 5981	5.0 0	0.5944	0 2762	-1 4509	-1.1783	-3 1328	50 4	-0.3802	3 -0 9677
10 0	-0 3251	-3 8573	10.0 0	-0 4965	0 0035	-1 6328	-1.1329	-1 8600	56.1	-0.2765	4 0.0837
15 0	-1 6861	-4 0842	15 0	0 0944	-0.1783	-1 5873	-1 0873	-1.2237	62 5	-0.1381	5 -1 0990
25 0	-1.3946	-4 6999	24.0 0	-0 0874	-0 3601	-1 5419	-1 1329	-0.9964	69 7	-0.1727	6 0 0837
35.0	-1.2649	-3 8573	33 0	-0 3105	-0 4420	-1.4932	-1 3618	-0 9677	76 9	-0.2418	7 -0 2072
50 0	-1.3621		54 0	-0 5734	-0.7047	-1 3618	1 3978	-0.9677	83.4	-0 2765	8 -1 7277
56 0	-1.4918	-1.5566	65 0	-1.0029	-1.4872	-1 6256	-1 6948	-1 2451	89 4	-0.3456	9 -1.6151
65.0	-1.9454	-1 6214	78 5	-0 4148	-4.4622	-4 0471	-3.0785	-1 7640	95.4	-0 4148	10 0 5812
76 0	-0 5843	-4.5054	79 5	-98 6954	-120.4735	-6 9499	-3.9090	-1.8737	101 1	-0.3110	11 0 0743
79 0	-7 6812	-8 6858	80.5	-64.5659	-51 0683	-4.6809	-2.9733	-0 5170	---	BOTTOM---	12 -0 7423
80 5	-14 6464	-31.6987	81 3	-30 1551	139 2278	-3 8621	-2 3182	-1.6866	38.2	0 2425	
81 0	-8 8915	-17.9673	82 0	-14 9501	0.5590	-3.4879	-2.3182	-1.6164	43.4	0.2770	
82 0	-6 6926	-11 5349	84 0	-0 2363	-5 9207	-2 8094	-1.8971	-1 7334	50 4	0.1041	
84.0	-1.5930		87.0	-1.0784	-5 8973	-2.2247	-1.6866	-1 7802	56.1	0.0003	
87.0	-2.0843		89.0	-2.6569	-3 3608	-2.0092	-1 7840	-1 8403	62.5	-0.3456	
89 0	-2.6288	-4 8428	93 0	-1 4462	-1 4462	-1 4179	-0.6577	-1 8122	69 7	-0 5186	
93.0	-1 4872		96 0	-1 6713	-1 5306	-1 1083	-1.3336	-1 6713	76.9	-0 4148	
96 0	-0 3110	0 8997	100 0	-0 8830	-1 3053	-0 7140	-0 9674	-0 5732	83 4	-0.2073	
100.0	2.1797	-1 1759	96 0	0 3278	0 1588	0 2151	0 4122	0.2714	89 4	0 0695	
96 0	0 6576	0 4154	84 0	0.1870	0 2433	0 2714	0.6938	0 7220	95 4	-0.0343	
84 0	2 1797	-1 1759	73 5	0 0837	-0.5734	0 3465	0.4779	0 7407	101 1	-0 5878	
73 0	0 6576	0 4154	54.0	-0 3105	-0 3105	-0 1791	0 0837	0.2151			
50 0	0 8305	0 6576	33 0	-0 5734	-0 4420	-0 5734	-0.3105	0 0837			
35 0	0.8091	0 5174	24 0	-0 5874	-0 6328	-0 6328	-0 1783	0 1399			
25 0	0 3878	-1 9130	10.0	-0 4965	-0 5419	-0 8600	-0 3601	0 3465			
10 0	0 4526	0 3230	5.0	-0 4965	-0 5419	-0 8600	-0 4510	0 4779			
5 0	0 3878	0.4202	2 5	-0 5419	-0 5419	-0 9510	-0 3601	0 4779			
2 5	0 5498	0 2258									

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TABLE A25. RUN 218, BC1W6V, DELF=45,CMU=2 O, (BN/B)W=0.5

RUN SEQ				PROPLULSIVE		WING / CANARD	PRESSURES			PAGE 505	
218 1	CLAERO =	0 5608	CDAERO =	0 9208	DCLC =	0.9986	DWLC =	1.7329	BASEPR =	8 1924	
ALPHA	PTOT	PSTAT	0	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0 07	2146 98	2142 91	4 07	57 45	0 0	2 66	1.220	2.002	3 222	86.9677	0.388510E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR
0 0	3.29	-0 78	-0.84	0 00	0 01	0.03	1.16	-0.17		1.16	0.49

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
	BP=3 375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLDC	CP	NO. CP
0 0	-0.1384	-3 2750	0 0	-1 3613	-1 1075	-0.3460	1 0078	-0.1768	38.2	-0.4733	1 -0.7045
2 5	-0 9829	-2 6116	2.5	-0.4306	1.2616	0.5001	-0 3460	-0.8536	43.4	-0.6021	2 -0.9491
5 0	-0 8622	-1.6463	5 0	0.8386	0 9232	0.2463	-0.3460	-0.8536	50 4	-0.3445	3 -0 9491
10 0	-0.2591	-1 4051	10 0	-0 5152	0 5001	-0 0922	-0 3460	-0 5998	56.1	-0.1513	4 -0.4599
15 0	-0 8622	-1.2845	15 0	0 3309	0.3309	-0.2614	-0.3460	-0.5998	62.5	-0.0869	5 -0.7045
25 0	-0.8019	-1 1035	24.0	0 1617	-0 0076	-0.3460	-0 3460	-0 5152	69 7	-0.0869	6 -0 2153
35 0	-0 8622	-1 0432	33 0	-0.2153	-0.4599	-0.7045	-0.7045	-0 9491	76 9	-0.1513	7 0.3264
50.0	-1.1035		54 0	-0 7045	-0 9491	-0 9491	4.1876	-0 9491	83 4	-0.2157	8 -0.6695
56 0	-1.2845	-1 4654	65 0	-1.1816	-1.8899	-1 1816	-1.1816	-1.1816	89.4	-0.3445	9 -0.5122
65 0	-1.8275	-2.3100	78.5	1 2010	-6 9128	-3.8863	-1.7611	-2 5339	95.4	-0.4089	10 0.8505
76 0	-0 4400	-5 6880	79.5	*****	-221.9840	-5.3933	-1.4307	-2 5194	101.1	-0.2801	11 0 3264
79 0	-8 5834	-10 3328	80.5	*****	-61 6089	-3 6951	-1.4743	-0.5598	---	---BOTTOM---	12 -0.0929
80.5	-21 6353	-47 7190	81 3	-25 1622	285 2717	-3 1290	-1.3436	-2.3452	38.2	0.0418	
81 0	-12.4909	-25.7275	82 0	-3.4338	-1.1259	-2.9982	-1.3872	-2.0404	43.4	0.0418	
82 0	-8 8332	-14 8860	84.0	-3 8691	-10 5314	-2.6499	-1 3872	-2 2146	50.4	-0.0869	
84 0	-1 8661		87.0	-0.4728	-8.1366	-1.6921	-1.4307	-2.3016	56 1	-0.5377	
87.0	-1 5613		89.0	-3 6046	-4.3383	-1.4556	-1.5606	-2 2943	62.5	-1.0528	
89 0	-2.9203	-4.5300	93.0	-1 4556	-1.6129	-1.2460	-0.4598	-2.2419	69.7	-1 1172	
93.0	-1 2460		96.0	-1.9799	-1.4556	-1.1411	-1.4032	-1.9799	76.9	-0.7308	
96 0	0 4926	0 7502	100 0	-2.9756	-3.1329	-0 9839	-1 3508	-0.4074	83.4	-0.3445	
100.0	5.9662	-2 1475	96.0	-0 0405	0 0643	0.5885	0 5361	0.2216	89.4	-0.0225	
96 0	0.4282	0 4282	84.0	0 2740	0 4836	0 8505	0.9030	0 5885	95.4	-0.2157	
84 0	5 9662	-2 1475	73 5	-0.2153	-0 7045	0 5185	0.5185	0 5185	101.1	-0.7308	
73 0	0 4282	0 4282	54.0	-0 7045	-0 7045	-0 4599	-0 2153	0 0293			
50 0	0 7502	0 6858	33 0	-1 4383	-0 7045	-0 9491	-0.4599	-0 4599			
35.0	0.7062	0 5855	24.0	-1.1921	-1 3613	-0.8536	-0 2614	-0 1768			
25 0	0 1029	-1 5862	10.0	-1 0229	-1 2767	-1.1921	-0.2614	-0 2153			
10 0	0.2236	0 1029	5 0	-0 9382	-1 1075	-1 4459	-0.2614	0.0293			
5 0	-0.0177	0.0425	2.5	-1.1075	-1.2767	-1.9536	-0.2614	0 0293			
2 5	0 1632	0 4045									

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TABLE A25 (Continued)

RUN SEQ 218 3  
 ALPHA 4 06  
 BETA 0 0  
 CLAERO = 0 4739  
 PTOT 2147 33  
 PSTAT 2143 38  
 Q 3 96  
 VEL 56 62  
 YAW 0.0  
 H/C 2 66  
 CMUC 1 257  
 CMUW 2.068  
 CMUT 3 325  
 CL 3 41  
 CD -0 63  
 CM -0.66  
 CROLL -0 02  
 CN 0.02  
 CY 0.03  
 CNTR 1.16  
 CMTR 0 03  
 DWLC = 1.8600  
 DCLC = 1 0782  
 BASEPR = 8.1924  
 HGT 87.0961  
 RN 0.383628E+06  
 CLTR 1.12  
 CDR 0.52  
 PAGE 507

\*\*\*\*\* CANARD \*\*\*\*\*  
 BP=3 375 BP=10 125  
 %X/C CP CP  
 0 0 -2 4881 -3 7909  
 2 5 -2 0538 -3 2946  
 5 0 -1 4335 -3 7287  
 10.0 -0 3169 -3 1705  
 15 0 -1 1854 -2 9223  
 25 0 -0 9992 -1.3715  
 35 0 -1 0613 -1 3094  
 50 0 -1 3094  
 56 0 -1.4335 -1 7436  
 65 0 -1.9297 -2 5502  
 76 0 -0 5030 -5 7759  
 79 0 -8 8155 -10 4906  
 80 5 -21 8523 -48 6767  
 81 0 -12 5379 -26 0180  
 82 0 -8 7763 -14 8664  
 84 0 -1 6114  
 87 0 -1 3875  
 89 0 -2 8551 -4 8416  
 93 0 -1 1332  
 96 0 0 5885 0 7872  
 100 0 7 0782 -2 2590  
 96.0 0 3898 0 3898  
 84 0 7 0782 -2.2590  
 73 0 0 3898 0 3898  
 50 0 0 6547 0 7209  
 35 0 0.7376 0 4895  
 25 0 0.1794 -1 8677  
 10 0 0 3034 0 0554  
 5 0 0 1174 0 1174  
 2 5 0 3655 0 3655

\*\*\*\*\* WING \*\*\*\*\*  
 BP = 2 BP = 6 BP = 12 BP = 16 BP = 22  
 %X/C CP CP CP CP  
 0.0 -1 0154 -0.8414 -1.0154 -0 0583 -1.7985  
 2 5 -0 4064 0.9859 0 0287 -0 5804 -1.8857  
 5 0 0 8988 0.8988 -0.1453 -0 4934 -1 3635  
 10 0 -0 3194 0 5508 -0.1453 -0.4934 -1 0154  
 15 0 0 5508 0 2897 -0.2323 -0 4934 -0 8414  
 24 0 0 2027 -0 0583 -0 4934 -0 5804 -0.6674  
 33 0 -0 5235 -0 5235 -1 0266 -1.0266 -1.0266  
 54 0 -0.7750 -1 0266 -1.0266 4.5077 -1.0266  
 65.0 -1 0008 -1 7955 -1.1332 -1.2657 -1.0671  
 78 5 1 6480 -6 6960 -3.9809 -1 6630 -1.7955  
 79 5\*\*\*\*\*-228 7833 -6 0446 -1 3875 -2.0144  
 80 5\*\*\*\*\*-68 2439 -4 2534 -1 3875 -0.6711  
 81 3-23.0166 281 0090 -3 4026 -1.4323 -1 7009  
 82.0 -0.1337 -0 0441 -3.2236 -1 3875 -1.7009  
 84 0 -3 7610 -10 2988 -2 7310 -1 4323 -1.7906  
 87 0 -0 2232 -8 1942 -1.8354 -1.4771 -1 7906  
 89 0 -3.3801 -4 3502 -1.6013 -1.7092 -1 8710  
 93.0 -1 0624 -1 4397 -1 3319 -0.5773 -1.9249  
 96.0 -1.6554 -1.3319 -1 1163 -1 4397 -1.9787  
 100 0 -2 4638 -2 5717 -1 0624 -1 3858 -0 4156  
 96 0 -0 4695 -0 4695 -0 1461 0 3390 0 2851  
 84 0 -0 2000 -0 4156 0 1234 0.8242 0 8242  
 73 5 -0 7750 -0 5235 -0.5235 0 2312 0.4827  
 54 0 -1 2781 -1 2781 -1.0266 -0.7750 -0 2719  
 33 0 -1.2781 -0 7750 -1.2781 -1.0266 -0 5235  
 24 0 -0 7544 -1 0154 -0 9284 -0.6674 -0 1453  
 10 0 -0 7544 -0 9284 -0 9284 -0.5804 -0.5235  
 5 0 -0 6674 -0 8414 -1 0154 -0 7544 -0.0204  
 2 5 -0 7544 -0.8414 -0.9284 -0 7544 0 2312

\*\*FUSELAGE\*\*  
 -----TOP-----  
 XLOC CP NO. CP  
 38 2 -0.6035 1 -0.7750  
 43.4 -0.6035 2 -1.5297  
 50.4 -0.4048 3 -1 0266  
 56.1 -0.2062 4 -0.7750  
 62 5 -0.0737 5 -1.0266  
 69 7 -0.2062 6 -0.5235  
 76.9 -0.2062 7 0.1774  
 83 4 -0.2724 8 -0.7929  
 89 4 -0.3386 9 -0 6312  
 95 4 -0.3386 10 0 6625  
 101.1 -0.3386 11 0.2851  
 ---BOTTOM---  
 12 -0.3078  
 38 2 0.1250  
 43 4 0.1912  
 50.4 -0.0737  
 56 1 -0.4711  
 62.5 -1.0008  
 69.7 -1.1995  
 76 9 -0.9346  
 83.4 -0.7359  
 89 4 -0 4048  
 95.4 -0.3386  
 101.1 -0.6697

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TABLE A25 (Concluded)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 509			
218 5	CLAERO =	0 8733	CDAERO =	1 0941	DCLC =	1 1664	BASEPR =	8 1924		
	ALPHA	PTOT	PSTAT	Q VEL	YAW	H/C	CMUC	CMUW	CMUT	
12 02	2147.26	2143.31	3 96	56.61	0.0	2.68	1.267	2 080	3.347	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	4 02	-0.04	-0.46	0.03	-0.00	0.01	1.70	0.24	1.57	0.79

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3 375	BP=10 125		BP = 2	BP = 8	BP = 12	BP = 16	BP = 22	----	TOP----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0.0	-7.4255	-3.9516	0 0	-1.2512	-0 9902	-1.6863	-1.4252	-5.4279	38.2	-0.9094	1	-0.7498
2 5	-7.3636	-4.0757	2.5	-0.5552	0 4020	-1.5993	-1.2512	-3.6875	43.4	-0.8432	2	-1.2529
5.0	-7.2395	-4 0757	5.0	0 4020	0.3150	-1.7732	-1.2512	-2.4694	50.4	-0.6782	3	-1.2529
10.0	-0.2917	-4.3240	10.0	-0.7291	0.0539	-1.7732	-1.1642	-1.5993	56.1	-0.3796	4	0 0048
15 0	-1.6565	-4.3859	15.0	0.1410	-0.2071	-1 6863	-1.2512	-1.3382	62.5	-0.2472	5	-1.2529
25 0	-1.5323	-4.9443	24 0	-0.2071	-0.4682	-1 5123	-1.3382	-1.2512	69.7	-0.2472	6	-0 2467
35 0	-1.4083	-4.5100	33.0	-0.2467	-0 4983	-1.5045	-1.5045	-1.0013	76.9	-0.3134	7	-0.0670
50.0	-1.5323		54.0	-0 7498	-0 7498	-1.2529	5.2875	-0 7498	83.4	-0.3796	8	-1.4144
56 0	-1 7806	-2.0907	65.0	-1 1743	-1 9027	-1.7040	-2.1013	-1.5715	89.4	-0.5120	9	-1 3606
65.0	-2.3387.	-2.0907	78.5	1 5408	-5 3462	-4.2869	-3.4259	-2.4325	95.4	-0.4458	10	0.5799
76.0	-0 6018	-5.6265	79.5*****	-229.9664	-7 8555	-4.6313	-2.6162	101.1	-0.3796	11	0.3104	
79 0	-9.7209	-10 3413	80.5*****	-76.2797	-5.4822	-3.6460	-0.6458	----	BOTTOM----	12	-0.6060	
80.5	-22.8121	-48.9194	81.3-31.5437	281 6157	-4.3179	-2.8849	-2.4370	38.2	0.1502			
81 0	-13 3187	-26 8876	82.0	-7 2734	1.4141	-3.9595	-2.7952	-2.2578	43.4	0.2164		
82 0	-9 4674	-16 2741	84 0	-2 4370	-9.3779	-3.2879	-2.1683	-2 3475	50.4	-0.1147		
84.0	-2.0341		87 0	-0.8249	-8.7511	-2.2131	-1.7654	-2.3923	56.1	-0.3796		
87 0	-2.0341		89.0	-3.8940	-4.4867	-2.0614	-1.7917	-2.4386	62.5	-0.9094		
89.0	-3.4259	-6 0747	93.0	-1.6840	-1.7378	-1.4684	-0.6060	-2.1153	69.7	-1.0418		
93.0	-1.6377		96 0	-2.1691	-1.8458	-1.0371	-1.1449	-1.7378	76.9	-0.9756		
96.0	0.2164	0.8124	100.0	-1.8458	-2 2229	-0.6599	-0 8215	-0.5520	83 4	-0.6445		
100.0	7.7657	-2 3662	96.0	-0.1208	-0.1748	0.3104	0.5799	0.3643	89.4	-0.1809		
96.0	0.4151	0 2826	84 0	0 1487	-0.0670	0.4182	0 8494	0 9033	95.4	-0.3796		
84 0	7.7657	-2 3662	73.5	-0 2467	-0.4983	0.2564	0.7595	0 7595	101.1	-0.9756		
73.0	0 4151	0.2826	54.0	-0 7498	-0.7498	-0 4983	-0.2467	0 2564				
50.0	0 7462	0.7462	33.0	-1 2529	-0 4983	-1 2529	-0.4983	0 0048				
35.0	0 8249	0.4528	24 0	-1 1642	-1.1642	-1.2512	-0.6422	-0 0331				
25 0	0 3287	-2 2149	10 0	-0 9902	-1 1642	-1 2512	-0.8162	0 2564				
10 0	0 5147	0.2666	5.0	-0 9032	-1 0773	-1 3382	-0.9902	0.2564				
5.0	0.3907	0.3287	2.5	-0 9902	-1.0773	-1.3382	-1.1642	0.5080				
2.5	0.5147	0.2046										

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TABLE 26 RUN 240, BC1W6V, DELF=0, CMU=0, (BN/B)W=0 5

RUN SEQ 240 2  
 ALPHA -0 03  
 BETA -0 01  
 CLAERO = 0 1475  
 PTOT 2139 20  
 CL 0 15  
 CDAERO = 0 0409  
 PSTAT 0  
 VEL 157 73  
 CM 0 00  
 YAW 0 01  
 H/C 2 66  
 CROLL 0 00  
 DCLC = 0.0  
 WING / CANARD  
 PRESSURES  
 DWLC = 0 0  
 CMUC 0 0  
 CMUW 0 0  
 CMUT 0 0  
 CNTR 0.12  
 CMTR -0 13  
 BASEPR = 8.1914  
 HGT 86 9818  
 RN 0.103952E+07  
 CLTR 0 12  
 CDTR 0.07

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **			
%X/C	CP	BP=3 375	BP=10.125	%X/C	CP	BP = 2	BP = 6	BP =12	BP =16	BP = 22	----	TOP----	NO	CP
0 0	0 2661	0 2007	0 0	0 0	0.5662	0 5776	0 5204	0 5204	0 4975	38.2	-0.0882	1	-0 1573	
2 5	0.0455	0.0537	2.5	-0 1209	-0 1095	-0 1896	-0.2354	-0.1667	43.4	-0.0794	2	-0 0580		
5 0	0 0128	-0 0117	5 0	-0 0751	-0.1209	-0 1553	-0 1896	-0 2354	50 4	-0.0446	3	-0 1904		
10.0	-0 1588	-0 1261	10.0	-0 0980	-0.2011	-0 2240	-0 2240	-0 2240	56 1	-0.0446	4	-0 0911		
15 0	-0.1343	-0 1424	15.0	-0 1438	-0 1896	-0 2240	-0.1896	-0.2011	62.5	-0.0446	5	-0.1242		
25 0	-0 1343	-0 1343	24.0	-0.1438	-0 1667	-0 2125	-0 1553	-0.2011	69.7	-0.0707	6	-0 0911		
35 0	-0 1261	-0 1261	33 0	-0 1242	-0 1242	-0 1573	-0 1904	-0 1904	76 9	-0.0707	7	0 5332		
50 0	-0 1261	-0 1261	54 0	-0 1242	-0 1573	-0 1904	1.7627	1 4979	83.4	-0.0446	8	-0.0201		
56 0	-0 1261	-0 1261	65.0	-0 1666	-0 1840	-0.2363	-0.2625	-0 2451	89 4	-0 1056	9	0.0509		
65 0	-0.1915	-0 1588	78 5	-0 2363	-0 4629	-0 3584	-0 3235	-0 3148	95.4	-0 1579	10	0.3629		
76 0	-0 1179	-0 2731	79 5	-0 4023	-0.4200	-0 4200	22 8806	52.4750	101 1	-0.1230	11	0 2282		
79 0	-0 3794	-0 3794	80 5	-0 4082	-0.4200	-0 1782	-0 5379	-0 9330	---	BOTTOM---	12	-0 0059		
80 5	-0 3787	-0 3787	81 3	-0 4200	52 4750	-0 1900	-0 1428	-0.2666	38 2	-0.0446				
81 0	-0 3787	-0 3669	82 0	-0 3669	-0 8858	-0 1958	-0.2135	-0.1664	43 4	-0.0359				
82 0	-0 3787	-0 3610	84 0	-0 1546	-0 1900	-0 2725	-0 2548	-0 2017	50 4	-0.0533				
84 0	-0 1015	-0 1015	87 0	-0 1958	-0 2135	-0 3079	-0 2902	-0 2312	56.1	-0 0097				
87 0	-0 0720	-0 0720	89 0	-0 2258	-0 2471	-0 3109	-0 3464	-0 2542	62 5	-0.0184				
89 0	-0.1143	-0.1405	93.0	-0 1832	-0 1832	-0 2258	-0.1549	-0 2825	69.7	-0.0446				
93 0	-0 1143	-0 1143	96 0	-0 1407	-0 1478	-0 1619	-0.1762	-0 2329	76.9	-0.0446				
96 0	-0 0707	-0 0969	100 0	-0 0981	-0 0981	-0 0556	-0 0697	-0 1478	83.4	-0.0446				
100 0	2 1605	-0 0794	96 0	0 0863	0 1289	0 1502	0 1502	0.1360	89 4	-0.0272				
96 0	0 0774	0 1123	84 0	0 1430	0 1643	0 2140	0.1643	0 1218	95 4	-0 0010				
84 0	2.1605	-0 0794	73 5	0 0413	-0 0911	0 0744	0.0744	0 1075	101 1	-0 0620				
73 0	0 0774	0 1123	54.0	-0 0580	-0.0580	-0 0580	-0 0580	-0 0580						
50 0	0 1472	0 1384	33 0	-0 0580	-0.0911	-0 0911	-0 0911	-0 1242						
35.0	-0.0035	-0 0035	24.0	-0 1095	-0.1324	-0 1438	-0 1209	-0 1667						
25 0	-0.1751	-0 1915	10 0	0 0751	-0 1209	-0 1209	-0 1095	-0 1573						
10 0	-0 1751	-0.2078	5.0	-0 0522	-0 1209	-0 1438	-0 1095	-0 1242						
5 0	-0 2323	-0 2405	2 5	-0 0637	-0 0866	-0 1209	-0 1209	-0 1242						
2 5	-0 3957	-0 4284												

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TABLE A26 (Continued)

RUN SEQ 240 4      CLAERO = 0 3606    CDAERO = 0 0643    DWLC = 0.0      DWLC = 0.0      BASEPR = 8.1914      PAGE 576  
 ALPHA PTOT      PSTAT      Q    VEL      YAW    H/C    CMUC    CMUW    CMUT      HGT      RN  
 4.03 2139 13    2109 07 30.06 157 88    0.01 2 67    0 0    0 0    0 0    87.2797 0.103690E+07  
 BETA      CL      CD      CM      CROLL    CN      CY      CNTR    CMTR      CLTR    CDTR  
 -0 01    0 36    0 06    -0.08    0 00    -0 00    0 01    0 34    -0.09    0 33    0.09

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **			
BP=3 375    BP=10.125			BP = 2		BP = 6		BP =12		BP =16		BP = 22		-----TOP-----	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	0 3608	0 1810	0 0	0 4255	0 2651	0 0819	-0.0784	-0.2388	38.2	-0.1438	1	-0.2599		
2 5	-0 5461	-0 6360	2.5	-0 1357	-0 4105	-0 6854	-0.6625	-0 7999	43.4	-0.1176	2	0.0049		
5 0	-0 4317	-0 5951	5 0	-0 2617	-0.3876	-0 5708	-0.6052	-0 7197	50.4	-0 0828	3	-0 3261		
10 0	-0 1294	-0 4154	10.0	-0.1242	-0.3533	-0.4449	-0.4678	-0 5021	56.1	-0.0741	4	-0 0613		
15 0	-0 3582	-0 3908	15 0	-0.2617	-0 3304	-0.3991	-0 4105	-0.4449	62.5	-0.0828	5	-0.2599		
25 0	-0.2846	-0.2928	24 0	-0 2388	-0 2617	-0.3418	-0.3304	-0.3762	69.7	-0.1176	6	-0 0613		
35 0	-0 2111	-0 2193	33.0	-0 2268	-0.2268	-0.2930	-0.2930	-0 2930	76.9	-0.1002	7	0 5441		
50 0	-0 2029		54 0	-0 1937	-0.2268	-0 2268	1.4284	-0 2599	83.4	-0.0828	8	0.0262		
56 0	-0 1866	-0.1784	65 0	-0.2310	-0.2484	-0.3094	-0.3181	-0 3355	89.4	-0.1525	9	0 1752		
65 0	-0 2356	-0 2275	78.5	-0 2745	-0 5099	-0.3791	-0.3617	-0.3966	95.4	-0.1874	10	0.3242		
76 0	-0.1458	-0 3092	79 5	-0.4469	-0.4587	-0 4469	24.1698	52 4734	101.1	-0.1525	11	0 1114		
79.0	-0 4154	-0 4072	80 5	-0.4646	-0.4528	-0.2228	-0.5471	-0.9069	---BOTTOM---			12	-0.0021	
80.5	-0.4292	-0.4233	81.3	-0 4823	48 5312	-0.2346	-0.2110	-0.3525	38.2	-0.0218				
81 0	-0 4056	-0 4351	82 0	-0 4056	-0.8656	-0 2523	-0 2463	-0 2936	43 4	-0.0218				
82 0	-0 3702	-0 3702	84 0	-0 2169	-0.2582	-0.3230	-0 2936	-0 2817	50.4	-0.0566				
84 0	-0.1402		87 0	-0.2287	-0 2640	-0 3289	-0.2876	-0.3171	56.1	-0 0043				
87.0	-0 1579		89 0	-0.2575	-0 2788	-0.3284	-0.3355	-0 3284	62 5	0 0131				
89 0	-0 1699	-0 1787	93 0	-0 1937	-0 2078	-0 2362	-0.1653	-0.3497	69.7	0 0131				
93 0	-0 1525		96.0	-0.1582	-0.1653	-0 1653	-0.1866	-0.2930	76.9	-0 0130				
96.0	-0 1089	-0 1264	100 0	-0 1227	-0 1085	-0 0589	-0.0873	-0.1582	83.4	-0.0130				
100 0	2 0004	-0 0915	96.0	0 1043	0.1468	0 1681	0 1610	0 1256	89.4	-0 0043				
96 0	0.0828	0 1003	84.0	0.1610	0.1894	0.2249	0.1823	0 1185	95.4	0 0131				
84 0	2.0004	-0.0915	73 5	0.0711	-0.1275	0 1042	0.1042	0.1042	101.1	-0.0566				
73.0	0.0828	0.1003	54 0	0.0049	-0.0282	-0.0613	-0.0613	-0.0613						
50 0	0 1700	0.1526	33 0	0 0049	-0 1275	0 0049	0.0049	-0.0613						
35 0	0 0095	-0 0395	24.0	-0 0441	-0 0326	-0 0326	-0.0097	-0 0784						
25 0	-0 1131	-0.2601	10 0	0 0361	0 0361	0 0819	0.1162	0.0711						
10.0	-0.0886	-0 0967	5 0	0.0705	0 0475	0.1277	0.1735	0.2035						
5 0	-0 1213	-0 1131	2.5	0.0933	0.1162	0 2079	0.2880	0.3029						
2.5	0 0421	0 1075												

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TABLE A26 (Concluded)

RUN SEQ 240 10  
 ALPHA 12.07  
 BETA -0 01  
 CLAERO = 0 8610  
 PTOT 2138 92  
 PSTAT 2108 86  
 CD 0 19  
 CL 0.86  
 CDAERO = 0 1890  
 Q VEL 30.06 158 25  
 CM 0.01  
 CROLL 0 00  
 WING / CANARD DCLC = 0.0  
 H/C 2 67  
 CN -0 00  
 PRESSURES CMUC 0 0  
 CMUW 0 0  
 CMUT 0 0  
 CY 0.01  
 CNTR 0.86  
 CMTR -0 01  
 BASEPR = 8 1914  
 HGT 87.2350  
 RN 0.103082E+07  
 CLTR 0.83  
 CDTR 0.21  
 PAGE 580

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
%X/C	BP=3 375	BP=10.125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP
	CP	CP		CP	CP	CP	CP	CP	XLOC		
0 0	-3 8222	-2 0249	0.0	-0 4792	-1 6817	-3 7772	-3 4911	-1 4412	38.2	-0.2397	1 -0.4916
2.5	-2.0330	-1.8288	2.5	-0 1700	-1 1778	-1.9336	-3 1934	-1 2923	43 4	-0.1874	2 0 1704
5 0	-1 3712	-1 6490	5.0	-0.6510	-0 9030	-1 3038	-3.0673	-1.3038	50 4	-0.1351	3 -0 7896
10 0	-0 0396	-1.5101	10 0	-0 1586	-0 7312	-0 9831	-1.3381	-1 2236	56.1	-0.1351	4 0.1042
15 0	-0 8892	-1 4856	15 0	-0 5480	-0.6396	-0 8342	-0 7541	-1 1435	62 5	-0.1525	5 -0.8227
25 0	-0 6278	-1 2079	24 0	-0 4792	-0 5021	-0.6052	-0.5708	-1 0518	69.7	-0.2048	6 -0 0613
35 0	-0 4971	-0.7912	33.0	-0 4254	-0 4916	-0.5247	-0.4916	-1 0544	76.9	-0.1961	7 0 5582
50 0	-0 3827		54 0	-0 3261	-0 3592	-0 3923	0 8325	-0 7234	83 4	-0.1525	8 0 0901
56 0	-0 3418	-0 4317	65.0	-0 3355	-0 3617	-0.4314	-0.4140	-0 6842	89.4	-0.1961	9 0.2958
65.0	-0 3664	-0.3745	78.5	-0 3268	-0.5970	-0 4750	-0.4227	-0 7888	95.4	-0.1961	10 0 1681
76 0	-0 1948	-0.4072	79 5	-0 5235	-0 5235	-0.5058	22 3770	52 4834	101.1	-0.1612	11 -0.0163
79 0	-0.4807	-0.4889	80 5	-0 5412	-0 5177	-0 2817	-0.5825	-0.7713	---	BOTTOM---	12 -0 0305
80 5	-0 4941	-0 5530	81.3	-0 5471	36 0757	-0.3053	-0.2817	-0 6238	38 2	0.0567	
81 0	-0 4823	-0 4646	82.0	-0 4882	-0 8361	-0 3230	-0.3525	-0 6120	43 4	0.0567	
82 0	-0 4587	-0.3348	84 0	-0 2582	-0.3289	-0 3761	-0.3584	-0.6592	50.4	-0.0130	
84 0	-0 2051		87.0	-0 2935	-0 3112	-0.3820	-0.3348	-0.6769	56.1	0.0916	
87 0	-0 2346		89 0	-0 3213	-0 3355	-0 3852	-0.3852	-0.6973	62 5	0.1177	
89 0	-0 2135	-0.3007	93.0	-0 2504	-0 2433	-0.2859	-0 2078	-0 6618	69.7	0.1351	
93 0	-0 1874		96 0	-0 2078	-0.1866	-0 2007	-0.2220	-0 6476	76.9	0.1177	
96 0	-0 1438	-0.2571	100 0	-0 1724	-0.1085	-0 0873	-0 1298	-0.2007	83 4	0.0828	
100 0	1 7825	-0 2222	96 0	0 0972	0 1539	0 1610	0 1397	-0 0092	89.4	0.0741	
96 0	0 1003	0 0567	84 0	0.1965	0 2249	0.2390	0.1894	0 0617	95.4	0 0654	
84 0	1 7825	-0 2222	73 5	0 1042	-0 1606	0.1373	0.1373	0.0711	101.1	-0 0566	
73 0	0 1003	0 0567	54 0	0 1042	0 0711	0.0711	0.0380	-0 0282			
50 0	0 2223	0.1438	33.0	0 1373	-0.1937	0 1704	0.1704	0 0380			
35.0	0 0667	0 0013	24 0	0.1735	0 1850	0.2079	0.2308	0 1048			
25 0	0 0421	-0 4072	10 0	0.2995	0 3224	0.3682	0.4140	0.3029			
10 0	0 1320	0 0912	5 0	0.3567	0 4025	0 4827	0.5170	0 4353			
5 0	0 1075	0 1402	2 5	0 4483	0.4713	0.4942	0 5285	0 4684			
2 5	0 5650	0.5569									

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TABLE A27. RUN 241, BC1W6V, DELF=0, CMU=0.5, (BN/B)W=0 5

RUN SEQ 241 1  
 ALPHA 0.03  
 BETA -0.01  
 CLAERO = 0.2014  
 PTOT 2138.99  
 CL 0.38  
 PSTAT 2117.52  
 CD -0.66  
 CDAERO = 0.1088  
 Q 21 47  
 CM -0.22  
 VEL 133 81  
 CROLL 0 01  
 WING / CANARD  
 DCLC = 0.0444  
 H/C 2.67  
 CN -0 01  
 PRESSURES  
 DWLC = 0.1389  
 CMUC 0 255  
 CMUW 0.535  
 CMUT 0 790  
 CY 0.01  
 CNTR 0.23  
 CMTR -0.16  
 BASEPR = 8 1914  
 HGT 87.2365  
 CLTR 0.23  
 RN 0.868204E+06  
 CDTR 0.04  
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***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
BP=3 375 BP=10.125			BP = 2		BP = 6	BP =12	BP =16	BP = 22	-----TOP----				
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP	
0 0	0 2236	0 3151	0.0	0 4402	0.4082	0.4883	0.5524	0.5685	38.2	-0.1335	1	-0.2500	
2 5	-0.0052	-0 0052	2.5	0.0715	0.1837	-0.0247	-0 1049	-0.0728	43.4	-0.1335	2	-0.1573	
5 0	-0.1310	-0.1081	5 0	0.2639	0 0875	-0 0728	-0.1530	-0.1209	50.4	-0.0847	3	-0.2500	
10 0	-0 1310	-0 1996	10.0	-0.0247	-0.0728	-0 1850	-0.2011	-0.1690	56.1	-0.0481	4	-0.1573	
15 0	-0.2340	-0.2110	15.0	-0 0568	-0 1209	-0 2011	-0 2011	-0.1690	62.5	-0.0359	5	-0.2036	
25 0	-0.2340	-0.2225	24 0	-0 1209	-0 1850	-0 1850	-0.1530	-0.1369	69.7	-0.0725	6	-0.1573	
35.0	-0 1768	-0 1996	33.0	-0 1573	-0 2036	-0.2963	-0 2963	-0.2500	76.9	-0.0603	7	0.6736	
50 0	-0.2110		54 0	-0 2036	-0.2963	-0 2963	1.9746	-0.2036	83.4	-0.0359	8	-0.2499	
56 0	-0 2454	-0 2225	65 0	-0 2555	-0.3287	-0.3531	-0.3287	-0.3043	89.4	-0.0847	9	-0.1804	
65 0	-0 3026	-0 3712	78.5	0.6963	-1.1951	-0 6094	-0.4385	-0.3775	95.4	-0.1579	10	0.5545	
76 0	-0 1424	-0 6915	79.5	21 4527	25 2014	-3 1682	-3.0031	73.1638	101.1	-0 1335	11	0.4452	
79 0	-1 0803	-1 1146	80.5	-18 7895	-11 2100	-0 2867	-0 3693	-1 0876	---BOTTOM---			12	-0 0414
80 5	-16.5437	-22 3315	81 3	-2 3673	64 3519	-0.3693	-0.2536	-0.3032	38.2	-0.0603			
81 0	2.4380	7 5818	82 0	-0.0307	-0 7325	-0 3857	-0.2784	-0 2289	43.4	-0.0481			
82 0	1 6371	-0 3940	84 0	-3 3829	-1.5664	-0 4683	-0.3362	-0.2619	50.4	0.0251			
84 0	0 6298		87.0	-0.2949	-1.6985	-0.5013	-0.3362	-0.2784	56 1	-0.0115			
87 0	1.4059		89 0	-1.0841	-0 6273	-0.5280	-0.3790	-0.2797	62.5	-0.0237			
89 0	0 1960	-0 9022	93.0	-0.3194	-0 1009	-0.3591	-0 1705	-0.2698	69.7	-0 0359			
93 0	0.3790		96.0	-0.4088	-0.3989	-0 2598	-0 2003	-0 2400	76 9	-0.0603			
96.0	1 0502	0.2936	100 0	-1.2430	-1.2132	-0.1009	-0.0811	-0.1407	83.4	-0.0481			
100 0	3 3442	-1 3781	96.0	-0 0016	0 0282	0 1175	0.1374	0.1275	89.4	-0.0359			
96 0	-0 0969	-0 0359	84.0	0.0977	0.1076	0 1771	0.1473	0.1175	95.4	0 0007			
84 0	3.3442	-1 3781	73 5	-0 1109	0.1671	0.0281	0.0281	0.0745	101.1	-0.0725			
73 0	-0 0969	-0.0359	54.0	-0.1109	-0 1573	-0.1573	-0.1109	-0.1109					
50 0	0 0862	0 1106	33.0	-0 1573	-0.0182	-0 1573	-0 1109	-0.1573					
35.0	-0.0280	-0 0395	24 0	-0 1209	-0.1369	-0 0888	-0 0728	-0.0888					
25 0	-0 2225	-0 3483	10 0	-0.1049	-0 1369	-0 1049	-0 1049	-0 2036					
10.0	-0 1538	-0.2225	5 0	-0 1209	-0 1690	-0.1209	-0 0408	-0.1109					
5 0	-0 2568	-0.2797	2 5	-0 2171	-0 2492	-0.1530	-0.1209	-0.0646					
2 5	-0 4855	-0 4513											

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TABLE A27. (Continued)

RUN SEQ		PROPLULSIVE WING / CANARD PRESSURES										PAGE 585	
241	3	CLAERO =	0 4280	CDAERO =	0 1253	DCLC =	0.0644	DWLC =	0 1798	BASEPR = 8.1914			
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
4 08		2138 92	2117.44	21 47	133 85	0 01	2 66	0 265	0 550	0.815	87 0205	0.867703E+06	
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0 01		0 67	-0 65	-0 17	0 01	-0 01	0 00	0.47	-0 11	0.46	0.04		
***** CANARD *****				***** WING *****						**FUSELAGE**		**MISC.**	
		BP=3.375	BP=10.125	BP = 2	BP = 6	BP =12	BP =16	BP = 22	----TOP----				
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP	
0 0	0 3608	0 0863	0 0	0.5364	0 4883	0.0074	-0 6178	-0.8263	38 2	-0.2189	1	-0 2963	
2 5	-0.5770	-0 8858	2 5	0.0875	-0 1529	-0.6499	-1 0186	-1.0667	43.4	-0.1823	2	-0.0183	
5 0	-0 5198	-0.6799	5 0	-0 0087	-0 2171	-0.4736	-0.6339	-0.7461	50 4	-0.1335	3	-0.3890	
10 0	-0 0280	-0 5427	10 0	0 0394	-0 3133	-0 4576	-0 4896	-0 5697	56.1	-0.0847	4	-0 1109	
15 0	-0 4054	-0 5427	15.0	-0 1850	-0 3613	-0 4415	-0 4415	-0 4736	62 5	-0 0847	5	-0 3890	
25 0	-0 3368	-0 4169	24 0	-0.2171	-0 3453	-0 3613	-0.3453	-0.3934	69 7	-0.1213	6	-0 1573	
35 0	-0 3140	-0 3368	33 0	-0 2963	-0 2963	-0 3426	-0.3890	-0 3890	76.9	-0.1091	7	0 7927	
50 0	-0 3368		54.0	-0 2500	-0 2500	-0 3426	2.7158	-0 3426	83.4	-0.1091	8	-0 0612	
56 0	-0 3254	-0 3597	65 0	-0 2921	-0 3897	-0 3775	-0 3165	-0 3043	89.4	-0.2067	9	0.0480	
65 0	-0 3826	-0 4855	78 5	0 6840	-1 1462	-0 5239	-0 3897	-0.3775	95 4	-0.2067	10	0.4154	
76 0	-0 1538	-0 8058	79 5	18.7677	25 7933	-2 4662	-2.3423	73.2128	101.1	-0.1701	11	0 2665	
79 0	-1 1603	-1.2060	80 5	-15 2545	-13 2236	-0 3610	-0 4022	-1.1122	---BOTTOM---		12	-0.1009	
80 5	-12 6289	-22 9404	81 3	-1 7232	62 8044	-0 4105	-0 3197	-0 4270	38.2	-0.0481			
81 0	-1 1122	0 8692	82 0	-0 1546	-0 9306	-0 4270	-0 3444	-0.3444	43.4	-0.0359			
82 0	0 3573	0 1262	84 0	-3.1844	-1 6736	-0 4848	-0.3940	-0.3610	50 4	0.0495			
84 0	0 6132		87 0	-0 2866	-1 7892	-0 5096	-0 3940	-0 3857	56 1	0.0007			
87 0	1 2820		89 0	-1 0542	-0 5776	-0 4584	-0.4187	-0 3988	62.5	0.0007			
89 0	0 2082	-1 0852	93 0	-0.3393	-0 0811	-0.3293	-0 1903	-0.3889	69.7	-0.0115			
93 0	0 3912		96.0	-0 4088	-0 3293	-0 2400	-0 2400	-0.3393	76.9	-0.0237			
96 0	1 0134	0 1960	100 0	-1 1634	-1 1435	-0 1108	-0.1506	-0.1605	83 4	-0 0237			
100 0	4 3200	-1.4268	96 0	0 0579	0 0977	0 1275	0 1175	0 0877	89 4	-0 0115			
96 0	-0 0603	-0 0237	84 0	0 1672	0 1672	0 1870	0 1473	0 0877	95.4	0.0129			
84 0	4 3200	-1 4268	73 5	-0 1109	0 2598	0 0745	0 0281	0 0745	101.1	-0.0603			
73 0	-0 0603	-0 0237	54 0	-0 0183	-0 0183	-0 1109	-0.1109	-0.1109					
50 0	0 1350	0 1228	33 0	-0 0183	0 0745	-0 0646	-0 0646	-0 1109					
35 0	-0 0166	-0 0624	24 0	0 0074	-0 0568	-0 0247	0 0234	-0 0568					
25 0	-0 1538	-0 4283	10 0	0 0394	-0 0247	0 0394	0 1356	0 0281					
10 0	-0.0624	-0 0967	5 0	0 0074	-0 0087	0 1035	0.2158	0 1671					
5 0	-0 0738	-0 1081	2 5	-0 0087	-0 0247	0 2158	0 3440	0 3062					
2.5	0 1207	0 1549											

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TABLE A27. (Concluded)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 587					
241 5	CLAERO =	0 9571	CDAERO =	0 2288	DCLC =	0.1012	DWLC =	0 2529	BASEPR =	8.1914		
	ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12 06	2138 85	2117.71	21.13	132 85	0.01	2 67	0 269	0.556	0.825	0.825	87.2061	0.859718E+06
	BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
	-0 01	1.31	-0.52	-0 09	0.01	-0.01	0.00	1.02	-0.03	1.01	0.13	

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
%X/C	BP=3.975	BP=10.125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP	
	CP	CP		CP	CP	CP	CP	CP	XLOC			
0 0	-4 4864	-2 4297	0.0	0 2817	-0.7607	-3.4810	-3.8556	-1.6077	38.2	-0.3281	1	-0.6025
2.5	-2.1275	-1.9533	2.5	0.1677	-0.8585	-1 7706	-3.5135	-1 4123	43.4	-0.2785	2	0 1509
5 0	-1.5117	-2 0927	5.0	-0.4187	-0.7444	-1.2657	-2.7479	-1.3634	50.4	-0.2041	3	-0.7438
10 0	0 1501	-1.8835	10.0	0.0211	-0 7444	-1.0213	-0.9236	-1.3145	56.1	-0.1669	4	0 1038
15.0	-1.0817	-1.8951	15.0	-0 6304	-0 7119	-0.8585	-0.7933	-1.3960	62.5	-0.1669	5	-0.9321
25.0	-0.7795	-1 5117	24.0	-0.5978	-0.6304	-0.6630	-0.6630	-1.2820	69.7	-0.2165	6	-0.0846
35 0	-0 6169	-0.9887	33 0	-0.5083	-0.5554	-0.6025	-0.5554	-1.0733	76.9	-0.2165	7	0 7462
50.0	-0.5239		54.0	-0 3200	-0.4612	-0.5083	2.8348	-0 8850	83.4	-0.1669	8	-0.1114
56.0	-0.5006	-0 5123	65.0	-0.4645	-0.6009	-0.5265	-0.5141	-0.7496	89.4	-0.2165	9	0 3224
65.0	-0.5588	-0.5820	78.5	0.9488	-1.2579	-0.6752	-0.5389	-0.7992	95.4	-0.2289	10	0 3224
76.0	-0.1172	-0.8144	79.5	17.5744	23 2196	-2.8687	-2.6590	74.3908	101.1	-0.1793	11	-0.1417
79 0	-1.2560	-1.1514	80.5-10	2844	-5 3433	-0.4025	-0.4193	-0.9981			12	-0.1315
80.5	-13.2456	-23.3034	81.3	-1 2413	61.1366	-0.4276	-0 3689	-0 7128	38.2	0 0438		
81.0	-1.3084	1.0823	82 0	0.2267	-0 9981	-0.4444	-0.4025	-0.6625	43.4	0 0562		
82.0	0 1260	0.3190	84 0	-2.8100	-1.5852	-0.4947	-0.4193	-0.6793	50.4	0 1306		
84.0	0.6461		87.0	-0 2934	-1.6440	-0 4947	-0.3773	-0.7296	56.1	0 0686		
87 0	1.1830		89.0	-1.1002	-0.6663	-0.5351	-0.4645	-0.7067	62.5	0 0934		
89.0	0 0438	-0.9108	93.0	-0.3838	-0.1618	-0.3838	-0.1518	-0.7269	69.7	0 1058		
93 0	0 2422		96.0	-0 4443	-0.3939	-0.2829	-0.2728	-0.6461	76.9	0 1058		
96 0	0.8745	0.1182	100.0	-1 2011	-1.1506	-0.1618	-0.1820	-0.1114	83.4	0 0686		
100.0	4.6557	-1.3943	96.0	0.0601	0.1005	0.1307	0.1106	-0.0105	89.4	0 0686		
96 0	-0.0182	-0.0058	84.0	0.1711	0.2014	0.2215	0.1610	0.0399	95.4	0 0562		
84.0	4.6557	-1.3943	73.5	0 0096	0.2451	0 1509	0.1509	0.1038	101.1	-0.0554		
73 0	-0 0182	-0.0058	54.0	0.0567	0 0567	0.0567	0.0567	-0.0374				
50.0	0.1802	0.1554	33.0	0.1509	0.1038	0.1509	0.1509	0.0096				
35.0	0.0920	0.0107	24.0	0 1025	0.1351	0 1840	0.2166	0.1025				
25 0	0.0107	-0.6052	10.0	0.2003	0 2166	0.3632	0.4121	0 2922				
10.0	0 1269	0 0920	5.0	0 2492	0.2817	0.3795	0.5098	0.4334				
5.0	0.1501	0.1501	2.5	0.2980	0.3469	0.4283	0 4935	0 4805				
2 5	0.5568	0.5568										

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RUN SEQ 256 3  
 ALPHA 3 59  
 BETA 12 43  
 CLAERO = 21417 40  
 PSTAT = 21119 00298.34  
 CD = 70  
 CDAERO = \*\*\*\*\*  
 VEL = 202 744512  
 CM = 18760 30  
 DCLC = 0 0  
 YAW H/C = 43 -2.00  
 CN = -0 00  
 CMUC = 0 0  
 CY = 0 00  
 DWLC = 0 0  
 CMUW = 0 0  
 CNTR = -0.00-18760 30  
 CMUT = 0 0  
 CMTR = 0 0  
 BASEPR = 81 8694  
 HGT = 0  
 RN = -65 4495 0.550929E+07  
 CLTR = 20558.70  
 CDTR = \*\*\*\*\*

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	----TOP----	NO	CP	
0.0	2.3422	2.3422	0 0	0.9873	0.9873	0.9873	0.9873	0.9873	38.2	0.7855	1 -0.5155	
2 5	2.3422	2.3422	2.5	0.9873	0.9873	0.9873	0.9873	0.9873	43.4	0.7855	2 -0.5155	
5 0	2.3422	2.3422	5.0	0.9873	0.9873	0.9873	0.9873	0.9873	50.4	0.7855	3 -0.5155	
10.0	2.3422	2.3422	10 0	0.9873	0.9873	0.9873	0.9873	0.9873	56.1	0.7855	4 -0.5155	
15.0	2.3422	2.3422	15.0	0.9873	0.9873	0.9873	0.9873	0.9873	62.5	0.7855	5 -0.5155	
25 0	2.3422	2.3422	24 0	0.9873	0.9873	0.9873	0.9873	0.9873	69.7	0.7855	6 -0.5155	
35 0	2.3422	2.3422	33 0	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155	76.9	0.7855	7 1.8029	
50 0	2.3422	2.3422	54 0	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155	83.4	0.7855	8 1.8029	
56 0	2.3422	2.3422	65 0	0.7855	0.7855	0.7855	0.7855	0.7855	89.4	0.7855	9 1.8029	
65.0	2.3422	2.3422	78.5	0.7855	0.7855	0.7855	0.7855	0.7855	95.4	0.7855	10 1.8029	
76 0	2.3422	2.3422	79.5	1.4164	1.4164	1.4164	1.4164	1.4164	101.1	0.7855	11 1.8029	
79 0	2.3422	2.3422	80 5	1.4164	1.4164	1.4164	1.4164	1.4164	---BOTTOM---		12 1.8029	
80 5	1.4164	1.4164	81 3	1.4164	1.4164	1.4164	1.4164	1.4164	38.2	0.7855		
81 0	1.4164	1.4164	82.0	1.4164	1.4164	1.4164	1.4164	1.4164	43.4	0.7855		
82 0	1.4164	1.4164	84 0	1.4164	1.4164	1.4164	1.4164	1.4164	50.4	0.7855		
84 0	1.4164		87 0	1.4164	1.4164	1.4164	1.4164	1.4164	56.1	0.7855		
87.0	1.4164		89.0	1.8029	1.8029	1.8029	1.8029	1.8029	62.5	0.7855		
89 0	0.7855	0.7855	93.0	1.8029	1.8029	1.8029	1.8029	1.8029	69.7	0.7855		
93 0	0.7855		96.0	1.8029	1.8029	1.8029	1.8029	1.8029	76.9	0.7855		
96 0	0.7855	0.7855	100 0	1.8029	1.8029	1.8029	1.8029	1.8029	83.4	0.7855		
100 0	0.7855	0.7855	96.0	1.8029	1.8029	1.8029	1.8029	1.8029	89.4	0.7855		
96 0	0.7855	0.7855	84 0	1.8029	1.8029	1.8029	1.8029	1.8029	95.4	0.7855		
84 0	0.7855	0.7855	73 5	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155	101.1	0.7855		
73 0	0.7855	0.7855	54 0	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155				
50 0	0.7855	0.7855	33 0	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155				
35 0	2.3422	2.3422	24.0	0.9873	0.9873	0.9873	0.9873	0.9873				
25 0	2.3422	2.3422	10 0	0.9873	0.9873	0.9873	0.9873	-0.5155				
10 0	2.3422	2.3422	5 0	0.9873	0.9873	0.9873	0.9873	-0.5155				
5 0	2.3422	2.3422	2.5	0.9873	0.9873	0.9873	0.9873	-0.5155				
2 5	2.3422	2.3422										

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TABLE A28 (Continued)

RUN SEQ 256 4      CLAERO = 0 1397    CDAERO = 0.0376    DCLC = 0 0      DWLC = 0.0      BASEPR = 8.1914      PAGE 599  
 ALPHA    PTOT      PSTAT    0    VEL      YAW    H/C      CMUC      CMUW      CMUT      HGT      RN  
 0 01    2141 82    2111.98 29 83 160 07    0 01 2 67    0 0      0.0      0 0      87.1731 0.987642E+06  
 BETA      CL      CD      CM      CROLL      CN      CY      CNTR      CMTR      CLTR      CDTR  
 -0 01    0 14    0 04    -0 11    0 01    -0 00    0 01    0.12    -0.11    0 12    0 08

\*\*\*\*\* CANARD \*\*\*\*\*

%X/C	BP=3 375		BP=10.125	
	CP		CP	
0 0	-0 1774		-0 1692	
2 5	-0 1692		-0 1774	
5 0	-0 1692		-0 1692	
10 0	-0 1692		-0 1774	
15 0	-0 1692		-0 1692	
25 0	-0 1692		-0 1692	
35 0	-0 1692		-0 1692	
50 0	-0 1774			
56 0	-0 1692		-0 1774	
65 0	-0 1692		-0 1774	
76 0	-0 1692		-0 1692	
79 0	-0 1692		-0 1692	
80.5	-0 1592		-0 1533	
81.0	-0 1592		-0 1414	
82 0	-0 1711		-0 1651	
84 0	-0 1592			
87 0	-0 1711			
89 0	0 9901		0 9550	
93 0	0 9813			
96 0	0 9637		0 9550	
100 0	0 9725		0 9637	
96 0	0 9813		0 9637	
84 0	0 9725		0 9637	
73 0	0 9813		0 9637	
50 0	0 9637		0 9550	
35 0	-0 1774		-0 1692	
25 0	-0 1692		-0 1774	
10 0	-0 1692		-0 1774	
5 0	-0 1610		-0 1774	
2 5	-0 1692		-0 1774	

\*\*\*\*\* WING \*\*\*\*\*

%X/C	BP = 2		BP = 6		BP = 12		BP = 16		BP = 22	
	CP		CP		CP		CP		CP	
0 0	0 5489		0 5605		0 4682		0 5258		0 5028	
2.5	-0 1433		-0 1202		-0 1087		-0 1318		-0 1433	
5 0	-0 1318		-0 1548		-0 1895		-0 1664		-0 2356	
10 0	-0 1779		-0 2356		-0 2587		-0 2240		-0 2471	
15 0	-0 2125		-0 2125		-0 2817		-0 2240		-0 2356	
24 0	-0 1664		-0 2010		-0 2125		-0 2125		-0 2356	
33 0	-0 2020		-0 2020		-0 2687		-0 3020		-0 2687	
54 0	-0 2020		-0 2353		-0 2687		1 3658		-0 2353	
65 0	0 9901		0 9725		0 9989		0 9989		0 9989	
78 5	0 9901		0 9901		1 0076		0 9901		0 9901	
79.5	-0 4562		-0 4680		-0 4740		20 1518		53 8379	
80.5	-0 4799		-0 4799		-0 1830		-0 5274		-0 8184	
81 3	-0 4680		45 2504		-0 2127		-0 1948		-0 3077	
82 0	-0 3849		-0 8244		-0 2364		-0 2305		-0 1889	
84 0	-0 1592		-0 1889		-0 2067		-0 2839		-0 2067	
87 0	-0 2186		-0 2661		-0 3136		-0 3017		-0 1770	
89 0	-0 2662		-0 2662		-0 3662		-0 3591		-0 2805	
93 0	-0 2161		-0 2161		-0 2590		-0 2018		-0 1876	
96 0	-0 1804		-0 1661		-0 1804		-0 2018		-0 2590	
100 0	-0 1375		-0 1232		-0 0803		-0 0875		-0 1947	
96 0	0 0769		0 1126		0 1412		0 1341		0 1341	
84 0	0 1412		0 0840		-0 2090		0 1627		1 7850	
73.5	-0 0018		-0 1686		0 0649		0 0649		0 1316	
54 0	-0 1019		-0 1353		-0 1353		-0 1353		-0 1353	
33 0	-0 1019		-0 2020		-0 1353		-0 1686		-0 1686	
24 0	-0 1202		-0 1318		-0 1433		-0 1433		-0 1895	
10 0	-0 0856		-0 1202		-0 1087		-0 1433		-0 2353	
5 0	-0 1087		-0 1087		-0 0971		-0 0625		-0 2353	
2 5	-0 0971		-0 1087		-0 0510		-0 1548		-0 2353	

\*\*FUSELAGE\*\*

-----TOP-----	
XLOC	CP
38.2	0 9637
43.4	0 9725
50 4	0 9725
56.1	0 9637
62.5	0 9637
69.7	0 9637
76 9	0 9725
83.4	0 9813
89 4	0 9725
95.4	0 9813
101.1	0 9813
---BOTTOM---	
38.2	0 9725
43.4	0 9813
50.4	0 9901
56.1	0 9725
62.5	0 9813
69.7	0 9813
76 9	0 9901
83.4	0 9901
89 4	0 9813
95 4	0 9813
101.1	0 9813

\*\*MISC.\*\*

NO.	CP
1	-0 2353
2	-0 1353
3	-0 2687
4	-0 1686
5	-0 2353
6	-0 1686
7	-0 2161
8	-0 2018
9	-0 2018
10	-0 2018
11	-0 2018
12	-0 2018

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TABLE A28 (Continued)

RUN SEQ 256 7  
 ALPHA 4 02  
 BETA -0 01  
 CLAERD = 0 3291  
 PTOT 2142.10  
 CL 0 33  
 CDAERD = 0 0614  
 VEL 160.76  
 YAW 0 01  
 DCLC = 0.0  
 H/C 2 66  
 CN 0 00  
 PRESSURES  
 DWLC = 0 0  
 CMUC 0 0  
 CMUW 0 0  
 CMUT 0.0  
 CNTR 0 32  
 CMTR -0 13  
 PAGE 601  
 BASEPR = 8.1914  
 HGT RN  
 87.0905 0.989892E+06  
 CLTR CDTR  
 0.31 0 10

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **		
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	----TOP----	XLOC	CP	NO.	CP
0 0	-0 1825	-0 1825	0 0	0 2342	-0 2009	-0 5673	-0 5330	-0 4528	38.2	0 9839		1	-0 3229
2 5	-0 1825	-0.1743	2 5	-0 1780	-0 7276	-0 9338	-0 7963	-1 1513	43.4	0.9926		2	0 0082
5 0	-0 1825	-0 1743	5 0	-0 4299	-0 5101	-0 6475	-0 6590	-0 7849	50.4	1.0013		3	-0 3560
10 0	-0 1743	-0 1743	10 0	-0.1551	-0.4643	-0.6017	-0 5330	-0 5673	56.1	0 9752		4	-0 0249
15 0	-0 1743	-0 1743	15 0	-0 3727	-0.4071	-0.4872	-0 4986	-0 4528	62.5	0.9839		5	-0 2898
25 0	-0 1743	-0 1825	24 0	-0.3040	-0.3498	-0.3841	-0 3841	-0 3612	69.7	0 9839		6	-0.0912
35 0	-0 1743	-0 1661	33 0	-0.2898	-0 2898	-0 3229	-0.3560	-0 3229	76.9	0 9926		7	-0 2116
50 0	-0 1825		54 0	-0 2567	-0 2567	-0.2567	1 0344	-0.2898	83.4	0.9926		8	-0 2045
56 0	-0.1743	-0.1743	65.0	0.9926	0 9752	0.9752	0.9665	0.9752	89.4	1.0013		9	-0 1974
65 0	-0 1743	-0 1743	78.5	0 9839	0 9839	0.9752	0 9578	0.9752	95.4	1.0013		10	-0 1904
76 0	-0 1743	-0.1661	79 5	-0 4722	-0 4722	-0 4722	21.9435	53 4298	101.1	0.9839		11	-0.1904
79 0	-0.1743	-0 1580	80.5	-0 4840	-0 4722	-0 2482	-0.5724	-0.7080	---	---BOTTOM---		12	-0 1974
80 5	-0 1716	-0 1716	81 3	-0 4663	36 5669	-0 2482	-0 2601	-0 3720	38.2	0 9752			
81 0	-0.1598	-0 1775	82.0	-0.3956	-0 7552	-0 2836	-0 2836	-0 3013	43.4	0 9839			
82 0	-0.1657	-0 1657	84.0	-0 2070	-0 2306	-0.2188	-0 3190	-0.3308	50.4	0.9665			
84 0	-0 1716		87 0	-0.2365	-0.2718	-0 3484	-0 3249	-0 1834	56.1	0 9926			
87 0	-0 1657		89.0	-0.2755	-0 2897	-0 3535	-0 3606	-0 3748	62.5	0.9926			
89.0	0 9752	0 9839	93.0	-0.2187	-0 2258	-0 2613	-0 1974	-0.1904	69.7	0.9665			
93 0	0 9926		96 0	-0.1833	-0.1833	-0.1833	-0.2116	-0.3393	76.9	0 9839			
96 0	0 9839	1 0013	100 0	-0 1407	-0 1052	-0 0911	-0 1123	-0.1974	83.4	0.9839			
100.0	0.9839	1 0100	96 0	0.0934	0 1288	0.1501	0.1430	0 1005	89.4	1.0013			
96.0	0.9665	1.0100	84 0	0 1714	0.1005	-0 1974	0 1856	1 3418	95.4	0.9839			
84 0	0.9839	1 0100	73.5	0 0413	-0 1574	0.0744	0 1075	0 1737	101.1	0.9752			
73 0	0 9665	1 0100	54.0	-0 0249	-0.0912	-0.0580	-0 0580	-0.0912					
50 0	0 9839	0 9926	33 0	0.0082	-0 1574	0.0082	0.0082	-0 0912					
35 0	-0 1825	-0 1743	24 0	-0 0177	-0 0063	-0 0063	0 0052	-0 0979					
25 0	-0 1743	-0 1661	10 0	0 0510	0 0739	0 0853	0 0968	0.0744					
10 0	-0.1743	-0 1661	5 0	0 1540	0.1655	0.1884	0 2227	0.1737					
5 0	-0.1825	-0 1743	2.5	0 2342	0 2342	0.3029	0.2800	0 3061					
2.5	-0 1825	-0 1661											

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TABLE A28. (Concluded)

RUN SEQ			PROPLULSIVE WING / CANARD PRESSURES								PAGE 603
256 9	CLAERD =	0 7415	CDAERD =	0 1727	DCLC =	0 0	DWLC =	0.0	BASEPR =	8 1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.93	2142 17	2112.00	30.17	161 13	0 01	2.72	0.0	0 0	0 0	88 9847	0 990759E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	0 74	0.17	-0.20	0 00	-0.00	0.01	0.75	-0 20	0.72	0.21	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
BP=3.375 BP=10.125			BP = 2		BP = 6	BP =12	BP =16	BP = 22	-----TOP----				
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP	
0 0	-0.2191	-0.2191	0 0	-2.3072	-4 2694	-5.9119	-3.1628	-1.4403	38.2	1.0096	1	-0.5822	
2.5	-0 2110	-0 2110	2 5	-0 1741	-1.9422	-2 4896	-2.5125	-1.3376	43.4	1.0096	2	0.2094	
5 0	-0.2191	-0.2110	5.0	-1 0410	-1 5201	-1.7027	-2.3528	-1.3376	50.4	1.0096	3	-1.0439	
10 0	-0 2110	-0 2110	10.0	-0 1855	-1 0638	-1 2920	-2.2388	-1.2578	56.1	1.0096	4	0 1104	
15 0	-0 2029	-0 2110	15.0	-0 7444	-0.8585	-1 0524	-1.6684	-1.2692	62.5	1.0269	5	-0.9450	
25 0	-0.2029	-0.2029	24 0	-0 5619	-0 6304	-0.7786	-1.2350	-1.1095	69.7	1.0096	6	-0 0875	
35 0	-0.2110	-0.2110	33 0	-0.4503	-0 5162	-0 6811	-0.6152	-1 0109	76.9	1 0009	7	-0 2429	
50 0	-0 2110		54.0	-0.3513	-0 3843	-0 4832	0 9019	-0 7141	83.4	1.0096	8	-0.2358	
56 0	-0.2191	-0 2110	65.0	1.0096	0 9922	1 0183	1 0009	1.0096	89.4	1.0009	9	-0.2287	
65.0	-0 2191	-0.2110	78 5	0.9835	1 0009	1.0096	0.9922	0.9922	95.4	1.0009	10	-0.2358	
76 0	-0 2191	-0 2029	79.5	-0 5317	-0 5259	-0.5259	25.2176	53.2330	101.1	1 0009	11	-0.2358	
79 0	-0 2191	-0 2110	80.5	-0 5376	-0.5435	-0 3262	-0 6609	-0.7607	---BOTTOM---			12	-0.2287
80.5	-0.2205	-0 2029	81.3	-0 5552	36 9734	-0.3438	-0.2616	-0.6022	38.2	0.9835			
81.0	-0.2088	-0.1735	82 0	-0.4906	-0 7960	-0.3438	-0.3086	-0 6374	43.4	0.9922			
82 0	-0.1970	-0 2146	84.0	-0 2558	-0.2910	-0 2675	-0 3673	-0.6316	50.4	0.9835			
84.0	-0.1970		87.0	-0 2792	-0 3027	-0.3791	-0 3556	-0.2088	56.1	0.9922			
87.0	-0 1912		89 0	-0 3135	-0.3277	-0.3701	-0.4054	-0 6810	62.5	0.9922			
89 0	0.9835	1 0096	93.0	-0.2570	-0 2711	-0.2923	-0 2287	-0.2429	69.7	1.0009			
93 0	1.0096		96.0	-0 2146	-0.2004	-0 2146	-0 2711	-0 6315	76.9	1.0096			
96.0	1.0009	1.0009	100.0	-0 1581	-0.1227	-0 1227	-0.1934	-0.2287	83.4	1.0096			
100.0	0.9922	1.0096	96 0	0.1105	0 1529	0.1388	0.1176	-0.0379	89.4	1.0096			
96 0	1.0009	1.0096	84.0	0 2165	0.1458	-0.2429	0.1670	1 2058	95.4	1.0009			
84 0	0.9922	1.0096	73.5	0 1104	-0 1864	0 1764	0 1434	0.1434	101 1	1.0183			
73.0	1.0009	1 0096	54 0	0 1104	0.0774	0.0774	0.0444	-0.0875					
50.0	1.0096	0.9922	33.0	0 2094	-0 2194	0.1764	0.1764	0.0444					
35 0	-0.2110	-0.2110	24.0	0.2252	0.2024	0 2138	0 2366	0 0769					
25 0	-0.2029	-0 2110	10 0	0 3735	0 3963	0.4191	0.4191	0 3083					
10.0	-0 2029	-0 2110	5.0	0.4876	0 5104	0.5560	0.5560	0 4732					
5.0	-0.2110	-0 2029	2 5	0.6359	0 5560	0.5218	0 5560	0.5062					
2.5	-0 2191	-0.2110											

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RUN SEQ	257 2	CLAERO =	0 1788	CDAERO =	0 1403	DCLC =	0 0	DWLC =	0.1333	BASEPR =	8.1914	PAGE	604
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMJC	CMUW	CMUT	HGT	RN		
-0 07	2141 96	2113 93	28 03	154 53	0 01	2.67	0 0	0.518	0 518	87 2884	0.966370E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
-0 01	0 31	-0 36	-0 18	0 00	-0.00	0 01	0 20	-0 16	0 20	0 04			

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
%X/C	BP=3.375 CP	BP=10.125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XLOC	CP	NO	CP	
0 0	-0.2408	-0 2233	0 0	0 5483	0.5605	0 4991	0 5360	0 4623	38 2	0 9840	1	-0 2757	
2.5	-0 2320	-0 2320	2 5	-0 2132	-0 1641	-0.2255	-0.2378	-0 2255	43 4	0.9746	2	-0.1336	
5 0	-0.2320	-0.2233	5 0	-0 1272	-0 1887	-0 2255	-0 2378	-0.3483	50 4	0 9746	3	-0.2401	
10 0	-0.2233	-0 2320	10 0	-0 2132	-0 2378	-0.3115	-0.2992	-0.3115	56.1	0.9933	4	-0 1336	
15 0	-0 2320	-0 2233	15 0	-0 2009	-0 2501	-0 2869	-0 2869	-0 3115	62 5	0.9746	5	-0 2046	
25 0	-0 2233	-0 2146	24.0	-0 2009	-0 2378	-0 2623	-0 2501	-0 2746	69 7	0.9746	6	-0.1691	
35 0	-0 2233	-0 2320	33 0	-0 2401	-0 2401	-0 2757	-0 2757	-0.2401	76 9	0.9840	7	-0.2451	
50 0	-0 2233		54 0	-0 2757	-0 2757	-0 3111	0 5055	1 5353	83 4	0.9933	8	-0.2527	
56 0	-0 2233	-0 2320	65 0	1.0027	0 9933	0 9933	1.0120	1 0120	89.4	0.9746	9	-0.2527	
65 0	-0 2320	-0 2320	78 5	0 9933	0 9933	1 0120	1.0027	1 0027	95.4	0.9840	10	-0 2527	
76 0	-0 2320	-0 2233	79 5	34 0489	36 2362	-3 4987	17 0612	57 2275	101 1	0.9840	11	-0.2603	
79 0	-0.2233	-0 2496	80.5	-21 8523	-21 8141	-0 2871	-0 6095	-1 0394	---BOTTOM---			12	-0.2603
80 5	-0 2365	-0 2238	81 3	17 4405	57 2275	-0 3123	-0 2365	-0 3313	38 2	1.0120			
81 0	-0 2112	-0 2365	82.0	1 5211	-5 2246	-0 3376	-0 2744	-0 2618	43 4	1.0027			
82 0	-0.2238	-0 2112	84.0	-3 3217	-2 8917	-0 2491	-0.3503	-0 2871	50.4	0.9933			
84 0	-0 2175		87 0	4 6252	-1 2607	-0 4388	-0 3566	-0 2301	56 1	1 0027			
87 0	-0 2175		89 0	-2.3831	-0 5647	-0 4581	-0 4201	-0 3212	62 5	1.0027			
89 0	0 9933	1 0027	93 0	0 0516	-0 0168	-0 3364	-0.2527	-0 2451	69 7	1 0027			
93 0	0 9933		96 0	-0 2527	-0 2831	-0 2451	-0 2299	-0 2831	76.9	0.9933			
96 0	0 9933	0 9933	100 0	-1 2875	-1 2570	-0 1158	-0 1082	-0 2527	83 4	1.0027			
100 0	0 9933	1 0027	96 0	-0.0016	0 0212	0 1049	0.1277	0 1125	89.4	1.0027			
96 0	1 0027	1 0027	84 0	0 1125	0 0592	-0 2527	0 1505	0 4016	95 4	0.9933			
84 0	0 9933	1 0027	73 5	-0 1336	-0 2401	0 0795	0 0795	0.1505	101.1	0 9840			
73 0	1 0027	1 0027	54 0	-0 1336	-0 1336	-0 1336	-0 1336	-0 1336					
50 0	1 0120	1 0120	33 0	-0 1336	-0 2401	-0 1691	-0 1336	-0 1336					
35 0	-0 2320	-0 2320	24 0	-0 1395	-0 1641	-0 1641	-0.1395	-0 1887					
25 0	-0.2233	-0 2320	10 0	-0 1027	-0 1150	-0 1395	-0 0781	-0 1691					
10 0	-0.2233	-0.2320	5 0	-0 0658	-0 1518	-0 1027	-0.0658	-0 1336					
5 0	-0 2233	-0.2233	2 5	-0 0781	-0 1027	-0 0658	-0 0781	-0 0981					
2 5	-0.2233	-0 2233											

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TABLE A29 (Continued)

RUN SEQ 257 5    CLAERO = 0 3762    CDAERO = 0 1635    DCLC = 0 0    DWLC = 0.1669    BASEPR = 8.1914    PAGE 606  
 ALPHA    PTOT    PSTAT    0    VEL    YAW    H/C    CMUC    CMUW    CMUT    HGT    RN  
 4.05    2141.89    2113 63 28 25 155 29 0 01 2.66    0 0    0 511    0.511    87.1404 0.968135E+06  
 BETA    CL    CD    CM    CROLL    CN    CY    CNTR    CMTR    CLTR    CDR  
 -0 01    0 54    -0 32    -0.21    0 01    -0 00    0 01    0.41    -0.19    0.40    0.07

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
BP=3.375    BP=10.125			BP = 2		BP = 6	BP =12	BP =16	BP = 22	-----TOP-----		NO	CP	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP			
0 0	-0 2405	-0.2318	0 0	0.2220	-0 1313	-0 4724	-0 5334	-0.7527	38 2	1.0202	1	-0 3720	
2 5	-0.2405	-0 2318	2 5	-0 2410	-0.8136	-1 0938	-1.0938	-1.1790	43.4	1.0295	2	0 0154	
5 0	-0 2405	-0 2318	5 0	-0 4724	-0 6552	-0.6674	-0.7039	-0.8867	50.4	1.0202	3	-0 4072	
10 0	-0 2492	-0.2405	10 0	-0 2288	-0.5334	-0 6187	-0.6430	-0.6430	56 1	1.0110	4	-0 0550	
15 0	-0 2405	-0.2318	15 0	-0.4115	-0 4237	-0 5212	-0 5455	-0.5455	62.5	1.0110	5	-0.3720	
25 0	-0 2405	-0.2231	24 0	-0 3384	-0 3750	-0 4359	-0.4359	-0.4481	69.7	1.0017	6	-0 1254	
35 0	-0.2405	-0 2405	33 0	-0 3368	-0.3720	-0.4072	-0.3720	-0.3720	76.9	1.0017	7	-0 2738	
50 0	-0 2405		54 0	-0 3016	-0 3368	-0 3720	0 3677	-0.3368	83.4	1.0017	8	-0 2587	
56 0	-0.2318	-0.2318	65 0	1 0202	1.0017	0.9924	1.0017	0.9924	89.4	1.0110	9	-0.2512	
65 0	-0.2405	-0 2231	78 5	0 9924	1 0017	1.0017	1 0017	1.0017	95 4	0.9924	10	-0.2587	
76 0	-0 2492	-0.2405	79.5	34 0082	36.4222	-3 5952	21.4965	56.7995	101.1	1.0017	11	-0 2663	
79 0	-0 2405	-0 2318	80 5	-21 8073	-21.8452	-0 3467	-0.6853	-0 9111	---BOTTOM---			12	-0 2587
80 5	-0 2526	-0.2401	81 3	17.2320	44.2234	-0 3717	-0.3341	-0 4219	38 2	1.0110			
81 0	-0 2338	-0.2338	82 0	2.4253	-4.8368	-0 4031	-0 3467	-0 3404	43.4	1.0017			
82 0	-0 2401	-0.2401	84 0	-1 8017	-2 9432	-0.2777	-0.3906	-0.3780	50.4	1.0017			
84 0	-0 2275		87 0	5 5295	-1.2372	-0 4596	-0.3843	-0.2526	56.1	1.0202			
87 0	-0 2526		89 0	-2.6285	-0 6361	-0.4851	-0.4474	-0.4248	62.5	0.9831			
89 0	1.0017	1.0110	93 0	-0.0851	-0 0323	-0 3493	-0.2663	-0.2663	69.7	0.9924			
93 0	1 0110		96 0	-0.2663	-0.3040	-0 2738	-0.2738	-0.3719	76.9	0.9924			
96 0	1.0110	1 0017	100 0	-1 2852	-1.2550	-0 1379	-0.1531	-0.2587	83.4	1.0017			
100 0	0 9924	0 9924	96 0	0 0206	0.0507	0 1111	0.1036	0.0734	89 4	0.9924			
96 0	1 0110	1.0017	84 0	0.1488	0 0809	-0.2663	0 1639	0.4054	95.4	1.0110			
84 0	0 9924	0 9924	73 5	-0.0902	-0.2311	0.0859	0 0859	0 2268	101.1	1.0202			
73 0	1 0110	1 0017	54 0	-0.0198	-0 0550	-0.0550	-0 0550	-0.0902					
50 0	1.0110	1.0017	33 0	-0 0198	-0.2311	-0.0198	-0.0198	-0.0902					
35 0	-0 2405	-0.2405	24 0	0 0027	-0.0095	-0.0338	-0.0338	-0 0826					
25 0	-0 2405	-0 2405	10 0	0.0880	0.0758	0 1002	0.1002	0 0506					
10 0	-0 2492	-0 2405	5 0	0.1367	0 1611	0.1733	0.2098	0.2268					
5 0	-0.2405	-0.2318	2 5	0 2098	0 2342	0 3682	0.3438	0.3677					
2 5	-0 2405	-0 2231											

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TABLE A29 (Concluded)

RUN SEQ 257 7  
 ALPHA 11.65  
 BETA -0 01  
 CLAERO = 0.7438  
 PTOT 2141 96  
 CL 0 99  
 CDAERO = 0 3048  
 VEL 27 91 154 36  
 YAW 0.01 2 71  
 H/C 0 0  
 CMUC 0 0  
 CN 0 02  
 CROLL 0.00  
 WING / CANARD DCLC = 0 0  
 PRESSURES DWLC = 0.2501  
 CMUW 0.558  
 CMUT 0 558  
 CNTR 0 82  
 CMTR -0.26  
 BASEPR = 8.1914  
 HGT 88.5055  
 RN 0.962044E+06  
 CLTR 0.80  
 CDTR 0.18

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
%X/C	BP=3.375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XL0C	CP	NO.	CP	
0 0	-0 2805	-0 2805	0.0	-2 0550	-3 5098	-5.8034	-5 0514	-1 6357	38 2	0 9656	1	-0.5655	
2 5	-0 2717	-0.2805	2.5	-0 2793	-1 8700	-2.4002	-3 6578	-1 4138	43.4	0.9468	2	0 1475	
5 0	-0 2717	-0 2805	5.0	-1 0192	-1 3644	-1 6357	-2 8317	-1.4014	50.4	0.9656	3	-0.8864	
10.0	-0 2805	-0 2805	10.0	-0 2793	-1.0438	-1.3151	-1.4631	-1 3891	56.1	0.9562	4	0 0762	
15 0	-0.2805	-0 2717	15 0	-0 7355	-0 8465	-1 0561	-1 5371	-1 3151	62.5	0.9562	5	-0.9934	
25 0	-0.2805	-0 2717	24 0	-0 5753	-0 6616	-0.7726	-0 8095	-1.2288	69 7	0.9562	6	-0 0664	
35 0	-0 2893	-0.2805	33 0	-0 4943	-0 5655	-0 6368	-0.6368	-1 0647	76.9	0.9843	7	-0 3184	
50 0	-0 2805		54 0	-0 4229	-0 4229	-0.4943	0.3614	-0 7081	83 4	0.9843	8	-0 3184	
56.0	-0.2805	-0 2805	65 0	1 0031	0.9843	0.9750	0 9937	0 9843	89 4	0.9843	9	-0 3184	
65 0	-0 2805	-0.2893	78.5	0 9937	0 9750	0.9843	0 9937	0.9843	95.4	0.9843	10	-0.3184	
76 0	-0 2805	-0.2805	79 5	34.8483	37.2729	-3.8337	25 2320	57.5037	101 1	0.9937	11	-0 3108	
79.0	-0 2893	-0 2893	80.5	-22.1911	-22 2925	-0 4187	-0 7742	-0 9773	---BOTTOM---			12	-0.3108
80 5	-0 2855	-0 2601	81.3	17 2848	42.2420	-0.4886	-0.3362	-0 6599	38.2	0.9843			
81 0	-0 2918	-0 2855	82 0	3 1359	-4 5319	-0.4886	-0 4061	-0 6599	43.4	0 9937			
82 0	-0.2791	-0 2918	84 0	-0.3934	-3.0402	-0.3489	-0 4568	-0.6980	50.4	0 9843			
84.0	-0.2791		87 0	6 8936	-1 2947	-0.5394	-0.4632	-0.2981	56.1	0.9843			
87 0	-0 2918		89 0	-2 9080	-0 6927	-0.5323	-0.4941	-0 6927	62.5	0.9750			
89 0	0 9656	0 9750	93.0	-0.3872	-0 0892	-0.3872	-0.3108	-0 3032	69.7	0.9843			
93 0	0 9562		96.0	-0 3108	-0 3261	-0 2955	-0 3337	-0 6163	76.9	0.9937			
96 0	0 9562	0 9656	100 0	-1 2504	-1.2657	-0 1962	-0 2191	-0 3032	83.4	0.9843			
100 0	0 9656	0 9843	96 0	0.0559	0 0864	0 1093	0.0941	-0.0434	89 4	0 9843			
96 0	0 9937	0 9656	84 0	0 1934	0 1246	-0 3108	0.1705	0 4837	95.4	0.9750			
84 0	0 9656	0 9843	73 5	-0 0664	-0.2803	0 1475	0 1475	0 1831	101 1	0 9562			
73 0	0 9937	0 9656	54.0	0 0762	0.0762	0.0762	0 0405	-0.0664					
50 0	0.9843	0 9656	33.0	0 1831	-0 2803	0 1475	0 1475	0 0049					
35 0	-0 2805	-0 2893	24.0	0 1646	0 1769	0 1893	0.2016	0 0536					
25 0	-0 2717	-0 2893	10 0	0 3866	0 3742	0 3866	0.3989	0 2901					
10 0	-0 2805	-0 2893	5 0	0 4729	0 4729	0.4975	0.5222	0.4327					
5 0	-0 2805	-0 2805	2 5	0 5962	0.5592	0.5222	0 5222	0 5040					
2 5	-0 2805	-0.2805											

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RUN SEQ	258 1	CLAERO =	0 1478	CDAERO =	0.3242	DCLC =	0.0	DWLC =	0.2681	BASEPR =	8.1914
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0 01	2142 38	2126.45	15.93	116.37	0 01	2 66	0.0	1 037	1.037	87.1184	0.728532E+06
BETA	CL	CD	CM	CROLL	CN	CY	CMTR	CLTR	CDTR		
-0 01	0 42	-0 68	-0 23	0 00	-0 00	0 01	0 22	-0 18	0.22	0.03	

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3	375	BP=10.125
%X/C	CP	CP	CP
0 0	-0 2588	-0 2434	
2 5	-0 2588	-0 2434	
5 0	-0.2588	-0.2434	
10 0	-0 2434	-0 3204	
15 0	-0 2588	-0 3204	
25 0	-0 2434	-0.2896	
35 0	-0.2434	-0 2588	
50 0	-0.2588		
56 0	-0 2434	-0 2588	
65 0	-0 2434	-0 2434	
76 0	-0 2588	-0 2434	
79 0	-0 2588	-0.2588	
80 5	-0 2190	-0 2079	
81 0	-0 2412	-0.1967	
82 0	-0.2301	-0 2190	
84 0	-0 1856		
87 0	-0 1967		
89 0	1 0370	1.0370	
93 0	1 0370		
96 0	1.0205	1 0370	
100 0	0 9876	1 0370	
96 0	0.9876	1 0534	
84 0	0.9876	1 0370	
73 0	0.9876	1 0534	
50 0	1 0041	1 0534	
35 0	-0 2588	-0 2588	
25 0	-0 2588	-0 2434	
10 0	-0 2434	-0 2742	
5 0	-0 2434	-0 2588	
2 5	-0 2434	-0 2280	

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	0.5289	0 5505	0 4857	0 4641	0 5289
2 5	-0.2487	-0.1191	-0.2919	-0.2271	-0.2055
5 0	-0 1407	-0 1839	-0 2919	-0.2055	-0.3135
10 0	-0 2271	-0 3351	-0 3135	-0 2919	-0.3135
15 0	-0.2487	-0.3351	-0 2703	-0.2271	-0.3135
24 0	-0.2487	-0 2919	-0.2487	-0.2055	-0.2919
33 0	-0 2449	-0 3074	-0.3074	-0.3074	-0 2449
54 0	-0 3074	-0.3074	-0 4323	0 1298	-0 2449
65 0	1 0205	1.0205	1.0205	1.0534	1.0534
78.5	1 0205	1 0205	1 0534	1.0370	1 0534
79.5	60 6012	64.2924	-5 6787	37.6410	100.0502
80.5	-38 5679	-38 1010	-0.2635	-0.7639	-1.4421
81.3	31.8701	81 5814	-0 3302	-0.2523	-0 2635
82 0	3.0833	-8.6470	-0 3858	-0 2857	-0.2301
84 0	-5.9228	-5.4007	-0.2746	-0.3302	-0.2746
87 0	8 2761	-1 7201	-0.4414	-0 3302	-0.2079
89 0	-3.3448	-0 4545	-0.4946	-0 4410	-0.3474
93 0	0.5491	0.3618	-0.4277	-0 2672	-0.2672
96 0	-0 2136	-0 3474	-0.3206	-0.2537	-0 2939
100 0	-2 0334	-2 1137	-0.1868	-0 1333	-0 2672
96 0	-0.0664	-0 0664	0 0808	0 0942	0.0942
84 0	0.0674	0 0272	-0 2672	0 1343	0 2815
73 5	-0 2449	-0 2449	-0 0576	0.0049	0.1922
54 0	-0.1200	-0.1200	-0 1825	-0 1200	-0.1200
33 0	-0 1200	-0 2449	-0 1825	-0 1200	-0.1825
24 0	-0 1407	-0 1623	-0 1623	-0 0975	-0 1623
10 0	-0 0759	-0.1191	-0 1407	-0 0327	-0 1825
5 0	-0.0543	-0 1191	-0 0975	-0.0543	-0 1200
2 5	-0 0975	-0 1191	-0 0543	-0.0543	-0 1200

\*\*FUSELAGE\*\*

	-----TOP----	**MISC.**
XLOC	CP	NO CP
38 2	1 0370	1 -0.2449
43.4	1.0370	2 -0.1825
50 4	1.0205	3 -0.3074
56.1	1.0370	4 -0.1825
62.5	1.0205	5 -0 2449
69 7	1.0370	6 -0.1200
76.9	1.0205	7 -0.2805
83.4	1 0205	8 -0 2805
89.4	1 0041	9 -0 2672
95 4	1.0205	10 -0.2805
101.1	1.0205	11 -0 2805
	---BOTTOM---	12 -0.2805
38 2	1 0534	
43.4	1.0370	
50.4	1 0534	
56.1	1 0534	
62 5	1.0370	
69 7	1.0370	
76.9	1 0370	
83.4	1.0370	
89.4	1 0205	
95 4	1 0370	
101.1	1.0205	

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TABLE A30. (Continued)

RUN SEQ				PROPLULSIVE		WING / CANARD		PRESSURES		PAGE 611
258 3	CLAERO =	0 3138	CDAERO =	0 3774	DCLC =	0.0	DWLC =	0.3758	BASEPR =	8.1914
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUT	HGT	RN
4 01	2142 38	2128 03	14 35	110 36	0 01	2 67	0 0	1 154	1 154	87 1726 0 692496E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0 01	0 69	-0 71	-0 27	0 00	-0 00	0 01	0 42	-0.22	0 41	0 06

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
BP=3 375		BP=10 125	BP = 2	BP = 6	BP =12	BP =16	BP = 22	----TOP----		NO	CP	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	XLOC	CP			
0 0	-0 2962	-0 2621	0.0	0 1639	-0 1238	-0 2437	-0.3397	-0 4356	38.2	1.0215	1	-0 3834
2 5	-0 2791	-0 2621	2 5	-0 2917	-0 6274	-0 9871	-1 0351	-0 9392	43.4	1 0033	2	-0.1061
5 0	-0 2791	-0 2621	5 0	-0 5075	-0 5795	-0 7473	-0 7713	-0.7473	50 4	1.0033	3	-0.4527
10 0	-0 2621	-0 2621	10.0	-0 2677	-0 5315	-0 6753	-0 6274	-0 6274	56.1	0.9850	4	-0.1061
15 0	-0 2449	-0 2791	15 0	-0 4596	-0 4596	-0.5555	-0 5315	-0 5075	62.5	1.0033	5	-0 3834
25 0	-0.2621	-0 2621	24 0	-0 3636	-0 4116	-0 4596	-0 3876	-0 4356	69.7	1 0033	6	-0.1061
35 0	-0 2621	-0 2791	33 0	-0 3141	-0 3834	-0 3834	-0.4527	-0 3834	76 9	1 0215	7	-0 3041
50 0	-0.2621		54 0	-0 3141	-0 3834	-0 3834	0 1713	-0 3834	83 4	1 0398	8	-0.3041
56 0	-0 2621	-0 2621	65.0	0 9850	0.9850	1 0215	1.0215	1 0033	89.4	1 0033	9	-0.3041
65 0	-0.2449	-0 2621	78.5	1 0033	1.0033	1.0033	1 0215	1 0033	95.4	1.0215	10	-0.3041
76 0	-0 2791	-0 2621	79 5	68.1572	71.7132	-6 9839	35.3208	110.9315	101.1	1.0215	11	-0.3041
79.0	-0 2791	-0 2621	80 5	-43 0049	-42 1898	-0.3547	-0.7867	-1 6508	---BOTTOM---		12	-0.2892
80 5	-0 2682	-0 2435	81 3	35 2344	87.3406	-0.4040	-0 3300	-0 3916	38.2	1.0033		
81 0	-0 2558	-0 2558	82 0	4 2622	-9 1807	-0 4410	-0.3300	-0 3300	43.4	1 0033		
82 0	-0 2558	-0 2558	84.0	-4 7616	-5 9467	-0.2806	-0.4040	-0 3547	50.4	1.0215		
84 0	-0 2806		87.0	9 2371	-1.7496	-0 5151	-0.3916	-0.2558	56 1	1 0033		
87 0	-0 2435		89 0	-3 8843	-0 3783	-0 5417	-0 4972	-0.4378	62 5	0 9850		
89 0	1 0033	1 0033	93 0	0 5130	0 4685	-0 4080	-0 2892	-0 3041	69 7	1 0033		
93 0	0 9850		96 0	-0.2149	-0 3189	-0.3189	-0 3189	-0.3932	76.9	0.9850		
96 0	0 9850	1 0033	100 0	-2 2650	-2 4136	-0.1852	-0 2000	-0 2892	83.4	1.0033		
100 0	0 9850	1 0033	96 0	-0 0813	-0 0515	0 0376	0 0970	0 0525	89 4	1.0033		
96 0	1 0033	1 0033	84.0	0 0970	0 0079	-0.2892	0 1416	0 2159	95 4	1.0033		
84 0	0 9850	1 0033	73 5	-0 2447	-0 2447	0 0326	-0 0367	0 1713	101.1	1.0215		
73 0	1.0033	1 0033	54 0	-0 1061	-0 1061	-0.1061	-0 1061	-0.1754				
50 0	0 9850	1 0215	33 0	-0 0367	-0 2447	-0.1061	-0 0367	-0 1754				
35 0	-0 2791	-0 2791	24 0	-0 0518	-0 0998	-0 0759	-0 0518	-0 1238				
25 0	-0 2621	-0 2621	10 0	0 0440	0 0440	0 0440	0 0920	0 0326				
10 0	-0 2791	-0 2621	5 0	0 0920	0.0920	0 1160	0 1879	0 1713				
5 0	-0 2621	-0.2791	2 5	0.1879	0 2119	0 2359	0 2838	0 2406				
2 5	-0 2621	-0 2621										

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TABLE A30. (Concluded)

RUN SEQ				PROPLULSIVE		WING / CANARD	PRESSURES			PAGE 614	
258 6	CLAERO =	0 7241	CDAERO =	0.4656	DCLC =	0 0	DWLC =	0 5092	BASEPR =	8.1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.96	2142.38	2128 03	14.35	110.31	0 01	2 73	0 0	1 123	1.123	89.1785	0.693272E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	1.23	-0.54	-0.35	0 00	-0 00	0 02	0 88	-0.30	0.85	0.20	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	----	TOP----	NO.	CP
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP		
0 0	-0.3133	-0 2962	0.0	-2.1142	-4 2485	-5.4236	-3.1694	-1.5866	38.2	1.0033	1	-0.6607
2 5	-0 2962	-0 3133	2 5	-0 2917	-1 9703	-2.5459	-2.5219	-1.2749	43.4	0.9667	2	0.1713
5 0	-0 3133	-0.2962	5 0	-1.0351	-1 3468	-1 8504	-2.6898	-1.2749	50.4	0.9667	3	-1.2848
10 0	-0.2962	-0 2962	10.0	-0.3157	-1.0351	-1 4187	-2 3061	-1.2509	56.1	0.9667	4	0.0326
15 0	-0 2962	-0.2962	15.0	-0.7473	-0 8672	-1 1550	-1.8984	-1 2749	62.5	0.9667	5	-1 0767
25 0	-0.3133	-0 3133	24 0	-0.5795	-0.6994	-0.8193	-1.1550	-1.2030	69.7	0.9485	6	-0.1061
35 0	-0 2962	-0 2962	33 0	-0.5914	-0 5914	-0.6607	-0.6607	-1 1460	76.9	0.9485	7	-0.3486
50.0	-0.2962		54 0	-0 4527	-0.5221	-0 5914	0.0326	-0 7994	83.4	0.9667	8	-0.3337
56 0	-0.3133	-0 2791	65.0	0 9850	0.9850	1 0215	1.0033	0.9850	89.4	0.9667	9	-0 3189
65 0	-0 3133	-0.2962	78 5	1.0033	0 9850	0.9850	0.9850	0 9850	95.4	0.9850	10	-0.3486
76.0	-0.3133	-0.3133	79 5	68.2930	71.6993	-7.6131	35.7653	110.9036	101.1	0.9850	11	-0.3189
79.0	-0.3133	-0 3133	80.5	-43 0788	-43.6592	-0.5028	-0.8855	-1.6261	---	BOTTOM---	12	-0.3337
80.5	-0.3176	-0.3176	81.3	35.2101	75.3013	-0.5892	-0.3547	-0.7126	38.2	0.9667		
81.0	-0.3176	-0.3299	82 0	4.7437	-9.3416	-0.6015	-0.3670	-0.7126	43.4	0.9485		
82.0	-0.3176	-0.3052	84 0	-3.5395	-6 3785	-0 3916	-0.4164	-0.7250	50.4	0.9667		
84 0	-0.3547		87.0	9.8913	-1 9100	-0.6015	-0.4410	-0.3299	56.1	0.9667		
87 0	-0 3052		89.0	-4.1666	-0.5120	-0 6011	-0.5418	-0 7646	62 5	0 9667		
89.0	1.0215	1 0033	93.0	0 4239	0.4387	-0.4526	-0 3486	-0.3337	69 7	0 9667		
93 0	0.9667		96 0	-0.2446	-0 4229	-0.3783	-0 3635	-0 7349	76.9	0.9850		
96 0	1 0033	0 9667	100.0	-2 2204	-2 3690	-0.2446	-0 2743	-0.3189	83.4	0.9850		
100.0	0 9850	0 9850	96.0	-0.0218	0 0228	0 0822	0 0674	-0.0663	89.4	1.0033		
96 0	0.9850	0 9850	84.0	0 1565	0.0822	-0.3337	0 1713	0.1416	95.4	0.9850		
84 0	0.9850	0 9850	73.5	-0 1061	-0.3141	0 1019	0 0326	0 1019	101.1	0.9850		
73 0	0.9850	0.9850	54.0	0.0326	0 0326	0.0326	0 0326	-0 1061				
50 0	0 9850	0.9850	33 0	0.1019	-0 3141	0.1019	0.1019	0.0326				
35.0	-0.3133	-0.2962	24.0	0 1879	0.2119	0 1879	0.2359	0 0440				
25.0	-0 2962	-0.3133	10.0	0.3798	0.3558	0 4277	0 4517	0.3099				
10 0	-0 3133	-0 2962	5.0	0.5236	0 4517	0.5476	0.5956	0.3793				
5.0	-0 2962	-0.3133	2.5	0 6435	0.5476	0.5716	0.5956	0.4486				
2 5	-0.2962	-0.2962										

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TABLE A31 RUN 259, BW6V, DELF=0, CMU=2 0, (BN/B)W=0 5

RUN SEQ	PROPLULSIVE										WING / CANARD			PRESSURES			PAGE 615	
259 1	CLAERO =	0 1194	CDAERO =	0 4941	DCLC =	0 0	DWLC =	0 4848	BASEPR =		8 1914							
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN							
-0 09	2142 45	2133 98	8 48	84 66	0 01	2 65	0 0	1 884	1 884	86 8020	0 533695E+06							
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR								
-0 01	0 60	-1 33	-0 31	0 00	-0 00	0 02	0.23	-0.22	0.23	0.02								

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3.375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	----	----	NO	CP
	CP	CP		CP	CP	CP	CP	CP	XLOC	CP		
0 0	-0.2711	-0 3001	0 0	0 5155	0 5561	0 4749	0 5155	0 4343	38.2	0.8482	1	-0.4060
2 5	-0 2422	-0 3001	2 5	-0.3779	-0 1748	-0 3372	-0 1748	-0 1748	43.4	0.8482	2	-0 1712
5 0	-0 2422	-0 3001	5 0	-0 2154	-0 2560	-0 2560	-0 1748	-0 3372	50 4	0 8482	3	-0 4060
10 0	-0 2711	-0 2711	10.0	-0 3372	-0 2966	-0 3779	-0.2560	-0.3372	56 1	0.8791	4	-0 1712
15 0	-0 2711	-0 3001	15 0	-0 2966	-0 2560	-0 3372	-0.2154	-0 3779	62.5	0.8482	5	-0 2887
25 0	-0 3001	-0 3001	24 0	-0 2560	-0 2966	-0 3779	-0 1748	-0.4185	69.7	0.8173	6	-0.1712
35 0	-0 2711	-0 3001	33 0	-0 2887	-0 4060	-0 4060	-0 2887	-0 4060	76.9	0.9100	7	-0 2802
50 0	-0.3001		54.0	-0 4060	-0 4060	-0 4060	-0.6408	-0 2887	83.4	0.8791	8	-0 3053
56 0	-0 2711	-0 2711	65.0	0.9718	0 9100	0 9100	0 9100	0.8791	89.4	0 8482	9	-0.3053
65 0	-0 3001	-0 2711	78 5	0 9409	0 9100	0 9409	0 9100	0 9100	95 4	0.9100	10	-0 2802
76 0	-0 3001	-0 3001	79 5	114 7948	121 8767	-11 9777	57 6682	187.1151	101.1	0 8791	11	-0 3053
79 0	-0 3001	-0 3001	80.5	-72 7190	-73 5979	-0 3559	-1.0038	-2 5924	----	----	12	-0 3304
80 5	-0.2723	-0 2932	81 3	60 9501	136 8709	-0 4813	-0.3349	-0 3768	38.2	0.8791		
81 0	-0.2723	-0 3141	82.0	6 0402	-16 4712	-0 4813	-0.3559	-0 2932	43.4	0.8482		
82 0	-0 2723	-0.3141	84 0	-9.8037	-9.8665	-0.3559	-0 4395	-0.3141	50 4	0 8791		
84 0	-0 3141		87 0	11 8928	-2 7178	-0 5859	-0.4186	-0 3559	56 1	0 8482		
87 0	-0.2514		89 0	-4 3300	0 0217	-0 5569	-0 4813	-0 3556	62 5	0 8482		
89 0	0.8482	0 8173	93 0	1 1034	1.1537	-0 4311	-0 3053	-0 2802	69.7	0.8791		
93 0	0 8482		96 0	-0 0789	-0 3807	-0.3053	-0.3053	-0 2802	76.9	0.8791		
96 0	0 9100	0 8791	100 0	-3 4999	-3 7264	-0 2047	-0.1543	-0 3053	83 4	0.8482		
100 0	0 8791	0 8791	96 0	-0 2047	-0 1796	0 0469	0.0972	0 0720	89.4	0 8482		
96 0	0 9100	0 8173	84 0	0 0469	-0 0286	-0 3053	0 1475	-0 5065	95.4	0.8791		
84 0	0 8791	0 8791	73 5	-0 1712	-0 4060	-0 0538	-0 0538	0 1810	101.1	0.8482		
73 0	0 9100	0.8173	54.0	-0 1712	-0 1712	-0.1712	-0 1712	-0 1712				
50 0	0.8791	0 8482	33 0	-0 1712	-0 4060	-0 1712	-0 1712	-0 1712				
35 0	-0 2711	-0 3001	24 0	-0 0936	-0 2560	-0.2560	-0 2154	-0 2966				
25 0	-0 2711	-0 3290	10 0	-0 1748	-0 2560	-0.2560	-0 1748	-0 2887				
10 0	-0 3001	-0.2711	5 0	-0 1342	-0 2154	-0.2154	-0 1748	-0 4060				
5.0	-0 2711	-0 2711	2 5	-0.1748	-0.1748	-0.1748	-0 1342	-0 4060				
2 5	-0 2711	-0 3001										

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TABLE A31 (Continued)

RUN SEQ 259 4  
 ALPHA 4 07  
 BETA -0 01  
 CLAERO = 0 3381  
 PTOT 2142 59  
 CL 1 03  
 PSTAT 2135.59  
 CD -1 55  
 CDAERO = 0 4439  
 Q 7 01  
 CM -0 38  
 PROPLULSIVE VEL 76 89  
 CROLL 0 00  
 WING / CANARD YAW H/C 0.01 2 69  
 CN -0 00  
 PRESSURES CMUC 0.0  
 CMUJ 2.106  
 CMUT 2.106  
 DWLC = 0.6881  
 CMY 0.02  
 CNTR 0.46  
 CMTR -0.28  
 BASEPR = 8.1914  
 HGT 87.8526  
 RN 0.486399E+06  
 CLTR 0.46  
 CDTR 0.05

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **			
BP=3.375 BP=10 125			BP = 2 BP = 6 BP =12 BP =16 BP = 22					-----TOP----					
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO	CP	
0 0	-0 3319	-0.2618	0 0	0.2621	0 0657	-0 4255	-0.3764	-0.7202	38.2	0.9989	1	-0.4212	
2 5	-0.3319	-0 2969	2.5	-0.3272	-0 7693	-1.1621	-0 9658	-1.0640	43.4	1.0363	2	-0 1372	
5 0	-0 3319	-0 2969	5 0	-0 4746	-0.6219	-0.8184	-0.7202	-0.8675	50.4	0.9989	3	-0 4212	
10 0	-0 3319	-0.2969	10.0	-0 3272	-0.5728	-0 7693	-0.5237	-0.7693	56.1	0.9615	4	-0 1372	
15 0	-0.2969	-0 2969	15 0	-0 4255	-0 4746	-0.5728	-0.4746	-0.6219	62.5	0.9615	5	-0.4212	
25 0	-0.3319	-0 3319	24.0	-0 3764	-0 4255	-0.4255	-0.3764	-0.5237	69.7	0.9989	6	-0.2792	
35 0	-0 2969	-0 2969	33 0	-0 4212	-0 4212	-0 4212	-0 4212	-0 4212	76.9	0.9989	7	-0.3400	
50 0	-0.3319		54.0	-0.4212	-0.4212	-0 4212	-0.9892	-0.4212	83.4	0.9989	8	-0.3704	
56 0	-0 2618	-0 3669	65.0	0.9989	0.9989	1 0363	0.9989	0.9989	89.4	1.0363	9	-0.3704	
65 0	-0 2969	-0.3319	78 5	0.9989	0.9989	0 9989	0.9989	0.9989	95.4	0.9989	10	-0.3704	
76 0	-0.3319	-0 3319	79 5	139.7529	146.0943	-15.6625	62.1850	226.1765	101.1	0 9989	11	-0 3704	
79 0	-0 3319	-0.3319	80 5	-87.9976	-89 1364	-0.4170	-1.0491	-3.0465	---BOTTOM---			12	-0.4008
80.5	-0.3664	-0.3159	81.3	73.8648	147 9922	-0.6193	-0.4170	-0.4170	38.2	0.9989			
81 0	-0 2906	-0 3664	82 0	4 7154	-19.8094	-0.5434	-0.3917	-0.3159	43.4	1.0363			
82 0	-0 3159	-0 2906	84.0	-11.6680	-10 5050	-0 3664	-0.4170	-0.4929	50.4	0.9989			
84 0	-0 3159		87.0	10.8088	-3 1476	-0 6446	-0.4675	-0.2906	56.1	0.9989			
87.0	-0 3159		89.0	-5 9386	0.0556	-0 6137	-0.5530	-0.4616	62.5	0.9989			
89 0	0 9989	0 9615	93.0	1 3944	1.4857	-0.4616	-0.3400	-0 3400	69.7	0.9989			
93 0	0 9989		96.0	-0 0661	-0.4313	-0.3704	-0.4008	-0 4313	76.9	1.0363			
96.0	1 0363	0 9989	100.0	-4 1736	-4.5388	-0.3095	-0.2486	-0.3704	83 4	1.0736			
100.0	1.0363	0.9989	96 0	-0 2486	-0 2486	-0 0052	0 0556	0.0252	89.4	0.9989			
96 0	0 9989	0 9989	84 0	-0 0052	-0 0661	-0 3095	0.1165	-1.0397	95.4	0.9989			
84.0	1 0363	0 9989	73.5	-0 2792	-0.4212	-0.1372	-0 1372	0 1468	101.1	0.9615			
73 0	0 9989	0 9989	54.0	-0.1372	-0.1372	-0.1372	-0 1372	-0 1372					
50 0	0 9989	1 0363	33 0	-0 1372	-0.2792	-0.1372	-0.1372	-0.2792					
35.0	-0.3319	-0.2969	24 0	-0.0325	-0.0817	-0 0325	0.0165	-0 0817					
25 0	-0 3319	-0.3319	10.0	0 1148	0.0657	0.1148	0.1639	0.0048					
10 0	-0 2969	-0.2969	5 0	0 1639	0.0657	0.2130	0 2621	0.0048					
5 0	-0.2969	-0 3319	2.5	0 1639	0 2130	0.3112	0 4095	0.0048					
2.5	-0 3319	-0.2969											

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TABLE A31 (Concluded)

RUN SEQ 259 7      CLAERO = 0 6345    CDAERO = 0 7661    WING / CANARD    PRESSURES      DWLC = 0.9677      BASEPR = 8.1914      PAGE 619  
 ALPHA    PTOT      PSTAT      Q    VEL      YAW    H/C      CMUC      CMUW      CMUT      HGT      RN  
 12 01    2142 73    2134 82    7 91    81 68    0 01    2 73    0 0      2 130    2 130    89 2304    0.517228E+06  
 BETA      CL      CD      CM      CROLL    CN      CY      CNTR      CMTR      CLTR      CDTR  
 -0 01    1 60    -1 13    -0 44    0 00    -0 00    0 02    0.93    -0 35    0.91    0.20

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**			
%X/C	BP=3'375	CP	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	CP	XLOC	CP	NO.	CP
0 0	-0 3663	-0 3663	0 0	-1.9599	-4 0048	-3.7872	-3.2217	-1.6989	38.2	0.8330	1	-0 7625	
2 5	-0.3663	-0 3975	2 5	-0 3067	-2 0034	-2 6126	-2 5691	-1.3944	43 4	0.8661	2	0 1180	
5 0	-0 3354	-0 3975	5 0	-1 0897	-1.5683	-2.0905	-2 4820	-1 3944	50.4	0.7999	3	-1 2656	
10 0	-0 3354	-0 3975	10 0	-0 3502	-1.1333	-1 7424	-2 5256	-1 3944	56.1	0.8330	4	-0 0077	
15 0	-0 3354	-0 3975	15 0	-0 7852	-0 9593	-1 2203	-2 1775	-1 3508	62.5	0.8661	5	-1 1398	
25 0	-0 3354	-0 3975	24 0	-0 7418	-0 8287	-0 9157	-1.0897	-1.3073	69 7	0 8992	6	-0 1335	
35 0	-0 3975	-0 3975	33 0	-0 5108	-0 6367	-0 7625	-0 7625	-1 2656	76 9	0 8661	7	-0.4030	
50.0	-0 3975		54 0	-0 5108	-0.5108	-0 6367	-1.2656	-0 8883	83.4	0 8330	8	-0 4299	
56 0	-0 4284	-0 3975	65 0	1.0317	1.0648	1 0317	0 9323	0 8992	89.4	0 8661	9	-0 4030	
65 0	-0.3975	-0 3663	78 5	0 9986	1 0317	0.9986	0.9323	0 9323	95.4	0.8330	10	-0 4030	
76 0	-0 3975	-0.3975	79 5	123 8879	130 4749	-13 5595	58.0197	200 5031	101.1	0 8661	11	-0.3761	
79 0	-0 4594	-0 4284	80.5	-77 7920	-78.9789	-0 5020	-1.0843	-2.5400	---	---BOTTOM---	12	-0.4030	
80.5	-0 2556	-0 3004	81.3	65 0744	119.9239	-0 6587	-0 3452	-0 8380	38.2	0.8330			
81 0	-0 3004	-0 3004	82 0	6.2393	-17.2543	-0.5916	-0.3900	-0 7036	43.4	0.7999			
82 0	-0.3004	-0.2780	84 0	-10 6251	-10.2892	-0.4572	-0.4572	-0.8156	50 4	0.8330			
84.0	-0 3229		87 0	12.5327	-3 1896	-0 6587	-0.4796	-0.3452	56 1	0.8330			
87 0	-0.2780		89 0	-5 3893	-0 2143	-0 7534	-0 5916	-0.8612	62.5	0 8330			
89 0	0 8992	0 8330	93 0	1.2681	1 1603	-0 5916	-0.4030	-0 4299	69.7	0.7999			
93 0	0 8661		96.0	-0.1065	-0 4569	-0 4838	-0.4569	-0 8342	76.9	0.7999			
96 0	0 8330	0.8661	100 0	-3 6912	-3.9877	-0.4299	-0.3761	-0.4030	83.4	0.8330			
100 0	0.8661	0 8330	96 0	-0 1604	-0 1604	0 0283	0 0283	-0 2143	89.4	0.8330			
96 0	0 8661	0 8661	84 0	0 0822	0.0013	-0 4030	0 0552	-1 2385	95.4	0 8330			
84 0	0 8661	0 8330	73 5	-0 0077	-0 2593	0 1180	-0 0077	0 2438	101 1	0 8330			
73 0	0 8661	0 8661	54 0	0.1180	-0 0077	-0 0077	-0 0077	-0.1335					
50 0	0 8992	0 8330	33.0	0.1180	-0 2593	0 1180	0.1180	-0.0077					
35 0	-0.4594	-0 3975	24 0	0.1719	0 1719	0 1284	0 2154	-0 0456					
25 0	-0.4284	-0 4284	10 0	0 4330	0 3024	0 3895	0 4330	0 2438					
10 0	-0 3975	-0 3975	5 0	0.5200	0.4765	0 4765	0 5635	0.4954					
5 0	-0 3663	-0 4284	2.5	0.6505	0 4765	0 5200	0.5635	0 4954					
2 5	-0 3975	-0 3975											

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TABLE A32. RUN 262, BW6V, DELF=0, CMU=0, (BN/B)W=0.25

RUN-SEQ	262 6	CLAERO =	0.1499	CDAERO =	0 0370	DCLC =	0 0	DWLC =	0 0	BASEPR =	8 1914	PAGE	620
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
0 05	2142 03	2112 08	29 95	158 47	0.01	2 66	0 0	0 0	0.0	86 9061	0.101979E+07		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR		
-0.01	0 15	0 04	-0 11	0 01	-0 00	0.01	0 15	-0 11		0.15	0.07		

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **			
%X/C	BP=3.375 CP	BP=10.125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XLOC	CP	NO	CP	
0.0	-0.1817	-0.1817	0.0	0.5502	0 5387	0 4927	0 5272	0 5042	38.2	-0.0951	1	-0.2311	
2.5	-0.1817	-0.1817	2.5	-0.1509	-0 1509	-0 2199	-0 2314	-0 1854	43.4	-0.0689	2	-0.0982	
5.0	-0.1817	-0.1817	5.0	-0.1279	-0 1624	-0 2199	-0 2314	-0.2659	50.4	-0.0602	3	-0.2311	
10.0	-0.1817	-0.1817	10.0	-0.1509	-0 1969	-0 2889	-0.2429	-0 2773	56.1	-0.0602	4	-0.1314	
15.0	-0.1734	-0.1817	15.0	-0.1854	-0 2084	-0 2773	-0 2314	-0.2773	62.5	-0.0602	5	-0.1979	
25.0	-0.1734	-0.1899	24.0	-0.1854	-0 2199	-0 2314	-0.1969	-0 2199	69.7	-0.0951	6	-0.1314	
35.0	-0.1899	-0.1899	33.0	-0.1646	-0.1979	-0 2311	-0 2311	-0 2311	76.9	-0.0864	7	-0.1978	
50.0	-0.1817		54.0	-0.1646	-0 1979	-0.2311	-0 0982	-0 2311	83.4	-0.0689	8	-0.2049	
56.0	-0.1817	-0.1817	65.0	-0.2089	-0 2526	-0 2964	-0 2701	-0.2526	89.4	-0.1126	9	-0.2049	
65.0	-0.1899	-0.1817	78.5	-0.1498	-0 4888	-0 3664	-0 3926	-0 3314	95.4	-0.1739	10	-0.2049	
76.0	-0.1899	-0.1899	79.5	-0.4570	-0 4748	2.6374	4 0988	53 6165	101.1	-0.1476	11	-0.1978	
79.0	-0.1817	-0.1817	80.5	-0.4629	-0 4748	-0.2145	-0 2440	-0 3446	---BOTTOM---			12	-0.1978
80.5	-0.1612	-0.1671	81.3	-0.4570	53 6165	-0 2085	-0.2026	-0 3328	38.2	-0.0689			
81.0	-0.1671	-0.1671	82.0	-0.4038	-0.4097	-0 2381	-0.2322	-0 2500	43.4	-0.0252			
82.0	-0.1671	-0.1730	84.0	-0.1908	-0.2145	-0 2736	-0.2973	-0 2500	50.4	-0.0951			
84.0	-0.1671		87.0	-0.2322	-0 2855	-0 3505	-0.3150	-0.1967	56.1	-0.0339			
87.0	-0.1671		89.0	-0.2476	-0.2832	-0.3402	-0 3544	-0 2832	62.5	-0.0339			
89.0	-0.1651	-0.1739	93.0	-0.2049	-0 2191	-0.2476	-0 2049	-0 1978	69.7	-0.0514			
93.0	-0.1651		96.0	-0.1764	-0 1764	-0 1764	-0.1907	-0 2547	76.9	-0.0602			
96.0	-0.1651	-0.1739	100.0	-0.1408	-0 1123	-0.0838	-0.0838	-0.1907	83.4	-0.0602			
100.0	-0.1651	-0.1651	96.0	0.0728	0 1226	0.1511	0.1440	0.1298	89.4	-0.0426			
98.0	-0.1739	-0.1739	84.0	0 1440	0.1084	-0 1978	0 1725	0.0942	95.4	-0.0164			
84.0	-0.1651	-0.1651	73.5	0.0348	-0 1646	0 0680	0 1012	0 1677	101.1	-0.1039			
73.0	-0.1739	-0.1739	54.0	-0.0982	-0.0982	-0.0982	-0 0982	-0.0982					
50.0	-0.1826	-0.1651	33.0	-0.0982	-0 1646	-0 1314	-0 1314	-0 1314					
35.0	-0.1899	-0.1817	24.0	-0.1165	-0 1394	-0.1394	-0.1165	-0.1739					
25.0	-0.1817	-0.1817	10.0	-0.0934	-0 1165	-0.1165	-0 0934	-0 1646					
10.0	-0.1899	-0.1817	5.0	-0.0705	-0 0934	-0.1050	-0 0360	-0 1646					
5.0	-0.1734	-0.1899	2.5	-0.0590	-0 0820	-0 0820	-0 0475	-0.1646					
2.5	-0.1817	-0.1817											

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TABLE A32. (Continued)

RUN SEQ	262 8	CLAERO =	0 3290	CDAERO =	0 0612	WING / CANARD	DCLC =	0 0	PRESSURES	DWLC =	0 0	BASEPR =	8.1914	PAGE	622
ALPHA	PTOT	PSTAT	0	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	87 4208	0.101824E+07		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	0 32	0 09			
	-0.01	0.33	0 06	-0.13	0 01	-0 00	0 01	0 33	-0.13						

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **		
%X/C	BP=3 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	XLOC	CP	NO	CP	
0 0	-0 1743	-0.1661	0 0	0 2342	0 0166	-0 1895	-0 2582	-0.3384	38 2	-0.0881	1	-0 3229	
2.5	-0.1580	-0 1661	2 5	-0 1551	-0 6131	-0.8307	-0.7963	-0 8536	43 4	-0.0620	2	-0 0249	
5 0	-0 1743	-0 1743	5 0	-0 4185	-0 5444	-0 6246	-0 6818	-0 7276	50.4	-0 0620	3	-0 3560	
10 0	-0 1743	-0 1743	10 0	-0 1666	-0 4528	-0 5444	-0 5444	-0.5444	56 1	-0.0707	4	-0 0580	
15 0	-0 1661	-0.1743	15.0	-0 3498	-0 3841	-0 4414	-0 4299	-0 4643	62 5	-0 0794	5	-0 3229	
25 0	-0 1743	-0.1580	24 0	-0 2811	-0 3154	-0 3727	-0.3612	-0 3841	69.7	-0 1317	6	-0 1243	
35 0	-0.1743	-0 1743	33.0	-0 2567	-0 2898	-0 3560	-0 3560	-0 3229	76 9	-0.1143	7	-0 2045	
50 0	-0 1661		54 0	-0 2236	-0 2567	-0 2898	-0 0912	-0 2898	83 4	-0 0881	8	-0 1974	
56.0	-0 1661	-0 1661	65 0	-0 2450	-0 2799	-0 3322	-0 3234	-0 3147	89 4	-0.1579	9	-0 1974	
65 0	-0 1661	-0 1661	78 5	0.2169	-0 4890	-0.3845	-0 4106	-0 3932	95 4	-0 1840	10	-0 1974	
76 0	-0 1661	-0.1661	79 5	-0 4251	-0.4663	2 5574	4 3021	53 4597	101.1	-0 1491	11	-0 1904	
79 0	-0 1661	-0 1580	80 5	-0 4486	-0 4545	-0 2482	-0 2718	-0 8494	---BOTTOM---			12	-0 1974
80 5	-0 1539	-0.1480	81 3	-0 4310	51 2608	-0 2188	-0 2188	-0 3484	38.2	-0.0533			
81 0	-0 1539	-0 1480	82 0	-0 3720	-0.7492	-0 2365	-0 2541	-0.2954	43.4	-0 0010			
82 0	-0 1657	-0.1598	84 0	-0 2129	-0 2129	-0 2482	-0 2954	-0.3131	50 4	-0 0794			
84 0	-0 1480		87 0	-0 2246	-0.2482	-0 3308	-0 3013	-0 1598	56.1	-0 0010			
87 0	-0 1598		89 0	-0 2684	-0 2967	-0 3606	-0.3677	-0.3535	62.5	0 0252			
89 0	-0 1579	-0.1579	93.0	-0 2045	-0 2329	-0 2471	-0 1974	-0 1904	69.7	0 0252			
93 0	-0 1579		96 0	-0 1691	-0.1762	-0.1833	-0.2045	-0 3180	76 9	0 0077			
96 0	-0 1666	-0 1666	100 0	-0.1407	-0 1052	-0 0768	-0.1052	-0 1904	83 4	-0 0010			
100 0	-0 1666	-0 1666	96 0	0 0934	0 1359	0 1572	0.1572	0 1005	89 4	0 0077			
96 0	-0 1666	-0 1666	84 0	0 1643	0 1147	-0.1974	0 1856	0 1714	95 4	0 0077			
84 0	-0 1666	-0.1666	73 5	0 0413	-0 1574	0.0744	0 0744	0 1737	101 1	-0 0794			
73 0	-0 1666	-0.1666	54 0	-0 0249	-0 0580	-0 0580	-0.0912	-0 0912					
50 0	-0 1579	-0 1666	33 0	-0.0249	-0.1905	-0 0580	-0 0249	-0 0912					
35.0	-0.1743	-0.1661	24 0	-0 0177	-0 0292	-0 0063	0.0166	-0 0635					
25 0	-0 1580	-0.1661	10 0	0.0853	0 0853	0 1082	0.1082	0.0744					
10 0	-0 1661	-0 1580	5 0	0 1540	0 1655	0 2113	0 2456	0 1737					
5.0	-0.1661	-0.1743	2 5	0 2227	0 2571	0 3143	0 2914	0 2399					
2.5	-0.1661	-0 1743											

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TABLE A32 (Concluded)

RUN SEQ	CLAERO =	0 7406	CDAERO =	0.1725	DCLC =	0 0	DWLC =	0.0	BASEPR =	8.1914
262 10	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUT	HGT	RN
ALPHA	2141 96	2111.78	30.17	159.41	0 01	2.72	0 0	0 0	89.0522	0.101814E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.74	0.17	-0.19	0.00	-0.00	0.01	0.76	-0 19	0.73	0.20

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3 375	BP=10.125
%X/C	CP	CP
0 0	-0.2110	-0 2110
2.5	-0 2029	-0 2110
5 0	-0.1948	-0 1866
10 0	-0 1948	-0 1948
15 0	-0 2029	-0.2029
25 0	-0.2029	-0 1948
35 0	-0.2029	-0 2029
50 0	-0 2029	
56.0	-0 1948	-0 2029
65 0	-0 2029	-0 2029
76 0	-0 2191	-0 2029
79.0	-0.2029	-0.2029
80 5	-0 2146	-0 1970
81 0	-0 2029	-0 1912
82 0	-0 1970	-0 1912
84 0	-0 1794	
87 0	-0.1970	
89 0	-0 1974	-0 1974
93 0	-0 2061	
96 0	-0 1887	-0 1974
100 0	-0 1887	-0 1887
96 0	-0.1974	-0.1887
84 0	-0.1887	-0 1887
73 0	-0 1974	-0 1887
50 0	-0 1974	-0 1974
35 0	-0.2110	-0.1948
25 0	-0.2029	-0 2029
10 0	-0 2110	-0 1948
5 0	-0 2029	-0.2029
2.5	-0.2029	-0.1948

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0 0	-1 9422	-3 8359	-5.5695	-3 2884	-1 4745
2 5	-0.1741	-1.8167	-2 5583	-2 5241	-1.2920
5.0	-1.0182	-1.3148	-1.8281	-2.6153	-1.3034
10.0	-0.1741	-0 9612	-1.2920	-2.1475	-1.2236
15.0	-0 6874	-0 8129	-1.0296	-1 6798	-1.2464
24 0	-0.5391	-0 6075	-0.7444	-1.2121	-1.1095
33 0	-0.4503	-0 5162	-0.5822	-0 8461	-0 9780
54 0	-0 3183	-0 3843	-0 3843	-0.0875	-0.6811
65.0	-0 3450	-0 3623	-0.4318	-0.4144	-0.6489
78 5	0 2281	-0.5360	-0.4405	-0.4231	-0.6923
79.5	-0.5082	-0 5259	2 5629	4.5769	53 2562
80 5	-0 5082	-0.5024	-0 3145	-0.3380	-0 8606
81 3	-0 5141	46.7670	-0 3027	-0.2440	-0 5905
82 0	-0 4261	-0 7079	-0 3086	-0 2969	-0 6374
84 0	-0.2381	-0 2851	-0 2969	-0 3556	-0.6551
87.0	-0 2616	-0.3203	-0.3732	-0.3497	-0.1912
89 0	-0.3065	-0 3347	-0.3630	-0.4195	-0.6527
93.0	-0 2429	-0 2499	-0 2711	-0 2358	-0.2358
96.0	-0.2004	-0 1863	-0 2075	-0.2499	-0.6315
100 0	-0.1510	-0 1227	-0 1227	-0 1793	-0 2287
96 0	0.1105	0.1458	0.1529	0.1246	-0.0521
84.0	0.2094	0 1529	-0.2287	0.1741	0 2094
73 5	0 1104	-0 1864	0.1763	0 1434	0.1763
54 0	0 0774	0 0774	0 0774	0.0774	-0 0545
33 0	0 1763	-0.1864	0.1763	0.1763	0.0444
24.0	0 2366	0.2252	0.2480	0.2708	0.0883
10 0	0.3963	0 4305	0.4077	0 4419	0.3413
5.0	0 5218	0 4990	0.5560	0.5788	0 4732
2.5	0 6473	0.5674	0 5446	0.5788	0.4732

\*\*FUSELAGE\*\*

----	TOP----	**MISC.**
XLOC	CP	NO. CP
38.2	-0.1019	1 -0 5492
43.4	-0.0932	2 0.2094
50.4	-0.1192	3 -1.0769
56.1	-0.1540	4 0.1434
62 5	-0.1974	5 -0.9120
69 7	-0.2495	6 -0.0545
76.9	-0.2234	7 -0.2429
83 4	-0.1626	8 -0 2287
89.4	-0.2321	9 -0.2429
95.4	-0.1887	10 -0.2287
101.1	-0.1453	11 -0.2358
---	BOTTOM---	12 -0.2358
38.2	0.0023	
43.4	0.0631	
50.4	-0.0324	
56 1	0.0805	
62.5	0.1413	
69 7	0.1760	
76.9	0.1413	
83.4	0 1065	
89.4	0.0805	
95.4	0.0631	
101.1	-0.0671	

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TABLE A33. RUN 265, BW6V, DELF=0, CMU=0 5, (BN/B)W=0.25

RUN SEQ	265 2	CLAERO =	0.1823	CDAERO =	0 1163	DCLC =	0 0	DWLC =	0 1278	BASEPR =	8.1904	PAGE	625
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
-0.04	2139.20	2121 91	17.29	120.66	0 01	2.66	0 0	0.495	0 495	86.9884	0.769511E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
-0 01	0.31	-0.36	-0.19	0 00	-0 00	0 00	0 21	-0.16	0.21	0.06			

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3:375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP-----	NO.	CP	
0 0	-0.2348	-0 2206	0 0	0.5228	0.5428	0.4830	0.5030	0 5030	38.2 -0.0903	1	-0.2079	
2 5	-0 2206	-0 2064	2.5 -0	2137	-0 1340	-0 3131	-0.2733	-0 2336	43 4 -0.0600	2	-0.0928	
5 0	-0 2206	-0 2064	5 0 -0	1738	-0 1539	-0.3131	-0.2534	-0.2534	50.4 -0.0448	3	-0.2655	
10 0	-0 2064	-0.2206	10.0 -0	2137	-0 2534	-0.3131	-0.3131	-0 2932	56.1 -0 0600	4	-0.1504	
15 0	-0 2206	-0 2348	15 0 -0	1937	-0.2336	-0 2932	-0 2733	-0 3131	62.5 -0.0600	5	-0 2079	
25 0	-0 2348	-0 2348	24.0 -0	2137	-0.2137	-0 2336	-0.2137	-0.2534	69.7 -0.1054	6	-0.1504	
35 0	-0.2064	-0 2206	33 0 -0	2079	-0.2079	-0.2079	-0.2655	-0.2079	76.9 -0.1054	7	-0.2573	
50 0	-0.2206		54 0 -0	2655	-0.2655	-0.2655	1 0006	-0.2079	83.4 -0.0751	8	-0.2450	
56 0	-0.2206	-0.2206	65.0 -0	3327	-0.6812	-0.3630	-0.3175	-0.2872	89.4 -0.1812	9	-0.2327	
65 0	-0 2206	-0 2064	78.5 1	3036	-0 9539	-0 4539	-0.4388	-0 3781	95.4 -0.2115	10	-0.2573	
76 0	-0 2206	-0.2064	79 5 55	7785	58.2279	8.1145	7.3146	92 3598	101.1 -0.1964	11	-0.2327	
79 0	-0 2064	-0 2206	80 5 -36	5876	-42.7276	-0 3320	-0.3320	-1 5518	---BOTTOM---	12	-0 2450	
80.5	-0.2090	-0 2294	81 3 26	3187	92.3598	-0 2704	-0.2192	-0.3217	38.2 -0.0751			
81 0	-0.2090	-0 1884	82 0 5	5107	-0 1987	-0 3217	-0 2910	-0 2807	43.4 -0 0297			
82 0	-0 2090	-0.1987	84 0 -0	3730	-2 4230	-0.3012	-0 3320	-0 2704	50 4 -0 0903			
84 0	-0 1884		87.0 10	5130	-1.9310	-0.3832	-0 3217	-0.2294	56 1 -0.0448			
87 0	-0 2192		89 0 -5	0671	-0 9850	-0.4176	-0 3807	-0.3067	62 5 -0 0448			
89 0	-0 2115	-0.2266	93.0 0	2730	-0 6396	-0 3190	-0 2573	-0 2327	69.7 -0 0448			
93 0	-0 2115		96 0 -0	1094	-0 7506	-0 2327	-0 2450	-0 2696	76.9 -0.0600			
96 0	-0 2115	-0 2266	100 0 -1	9469	-1 3179	-0 0970	-0 1094	-0.2450	83.4 -0 0600			
100 0	-0 2115	-0 2266	96 0 -0	0477	0 0386	0 1127	0 1250	0.1127	89.4 -0 0448			
96 0	-0 2115	-0 2266	84.0 0	1003	0 1003	-0 2573	0 1743	1 1732	95 4 -0 0297			
84 0	-0 2115	-0.2266	73 5 -0	1504	-0 2079	0.0798	0.0798	0 1949	101.1 -0 1206			
73 0	-0 2115	-0 2266	54 0 -0	0928	-0 0928	-0 0928	-0.1504	-0 1504				
50 0	-0 2115	-0 2266	33.0 -0	0928	-0 2079	-0 0928	-0 1504	-0 1504				
35 0	-0 2206	-0 1922	24.0 -0	1340	-0 1141	-0 1539	-0 1340	-0.1937				
25 0	-0 2064	-0 2206	10 0 -0	0942	-0 0942	-0 1340	-0 1141	-0.1504				
10 0	-0 2206	-0 2064	5 0 -0	0345	-0 0942	-0 0942	-0 0942	-0.0353				
5 0	-0 2206	-0 2206	2 5 -0	0345	-0 1141	-0 0743	-0.0743	-0 0928				
2 5	-0 2064	-0 2206										

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TABLE A33. (Continued)

RUN·SEQ 265 4  
 ALPHA 4 03  
 BETA -0 01

CLAERO = 0.3653  
 PTOT 2139 20  
 CL 0 53

PROPLULSIVE CDAERO = 0.1431  
 VEL 17.29 120 74  
 CM 0 00

WING / CANARD DCLC = 0 0  
 YAW H/C 0.01 2.98  
 CN CROLL 0 00

PRESSURES DWLC = 0.1613  
 CMUC 0 0  
 CMUW 0.494  
 CMUT 0 494  
 CY CNTR CMTR 0.40 -0.19

BASEPR = 8.1904  
 HGT 97.6337  
 CLTR 0.40

PAGE 627  
 RN 0.768174E+06  
 CDTR 0.09

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**			
BP=3.975 BP=10.125			BP = 2 BP = 6 BP =12 BP =16 BP = 22					-----TOP----					
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP	
0 0	-0.2206	-0 2206	0.0	0.1845	-0.0544	-0 3729	-0 2137	-0.5122	38.2	-0.0903	1	-0.3806	
2.5	-0 2206	-0 2206	2.5	-0 1738	-0 6913	-1 0297	-0.9899	-0 9501	43.4	-0.0600	2	0 0223	
5.0	-0.2206	-0 2064	5.0	-0 4326	-0.5122	-0.6316	-0.6515	-0 7710	50.4	-0.0600	3	-0.3806	
10.0	-0.2348	-0.2064	10 0	-0 1539	-0 4525	-0 5719	-0 5122	-0.5520	56.1	-0.0751	4	-0 0353	
15 0	-0 2206	-0.2064	15 0	-0.3729	-0 3729	-0.4923	-0 3729	-0 4525	62.5	-0.0903	5	-0.3230	
25 0	-0 2206	-0.2206	24.0	-0.2733	-0 2932	-0.3928	-0.3131	-0 3729	69.7	-0.1509	6	-0.0928	
35 0	-0.2206	-0.2064	33.0	-0 3230	-0 3230	-0.3806	-0.3806	-0 3806	76.9	-0.1509	7	-0.2327	
50 0	-0 2206		54.0	-0.3230	-0 3230	-0.3230	1 0581	-0.3230	83.4	-0.1206	8	-0 2080	
56.0	-0.2206	-0.2064	65 0	-0 3933	-0.7266	-0.4084	-0 3781	-0 3630	89.4	-0.2115	9	-0.2204	
65 0	-0.2206	-0.2064	78 5	1.4096	-0 9842	-0.4842	-0.4691	-0.4388	95.4	-0.2266	10	-0.2450	
76.0	-0 1922	-0 2206	79 5	55.9219	58.5356	7.2736	7.0688	92 3714	101.1	-0.1964	11	-0.2327	
79 0	-0 2064	-0.2206	80.5	-36.7102	-43 2707	-0.3627	-0 3422	-1 3877	---BOTTOM---			12	-0 2204
80.5	-0.2090	-0.2192	81.3	25 8577	92.3714	-0.2704	-0.2294	-0 3525	38 2	-0.0600			
81.0	-0.1884	-0 1884	82.0	6 3102	-0.3525	-0.2910	-0.2500	-0.3115	43.4	0.0006			
82 0	-0.2090	-0.1987	84 0	0 2318	-2 2488	-0.3012	-0.3217	-0.3012	50.4	-0.0751			
84.0	-0 2192		87.0	11 6917	-1.9413	-0 4037	-0.3422	-0.1884	56.1	0.0006			
87.0	-0.2090		89 0	-5.9427	-0 9973	-0 4053	-0 4053	-0 3807	62.5	0.0158			
89.0	-0 2115	-0.2115	93 0	0.4580	-0 6273	-0.2943	-0.2327	-0 2080	69 7	0.0309			
93 0	-0 2115		96.0	-0.1094	-0 7260	-0.2204	-0.2327	-0.3190	76.9	0.0006			
96.0	-0.1964	-0.2115	100.0	-1.9716	-1.3673	-0 1217	-0 1340	-0 2204	83 4	-0.0297			
100 0	-0.1964	-0 2115	96 0	-0 0107	0.0756	0.1373	0.1496	0 1250	89.4	0.0006			
96 0	-0.1964	-0 2115	84.0	0 1496	0.1250	-0.2204	0.1990	1.2595	95.4	0.0006			
84 0	-0 1964	-0 2115	73.5	-0 1504	-0 2079	0.1374	0.0798	0.3100	101.1	-0.1054			
73.0	-0 1964	-0 2115	54.0	-0 0353	-0.0353	-0 0353	-0 0928	-0.0928					
50 0	-0 2115	-0 2115	33 0	0.0223	-0 1504	-0 0353	-0.0353	-0 0928					
35 0	-0 2064	-0 2064	24 0	0 0451	0 0252	0.0053	0.0451	-0 0146					
25.0	-0 2206	-0 2064	10 0	0 1446	0 1446	0 1446	0 1845	0 0798					
10 0	-0 2064	-0 2206	5 0	0 2442	0 2243	0 2243	0 3039	0 1949					
5 0	-0 2348	-0 2206	2 5	0 3238	0 2840	0.3437	0.3835	0.2525					
2 5	-0.2064	-0 2206											

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TABLE A33 (Concluded)

RUN SEQ 265 6      CLAERO = 0 7406      CDAERO = 0 2860      DWLC = 0 2278      BASEPR = 8.1904  
 ALPHA PTOT      PSTAT      0 VEL      YAW H/C      CMUC      CMUW      CMUT      HGT      RN  
 12 04      2139 13      2121 73      17 40 121 23      0 01 2 73      0 0      0 501      0.501      89.2687      0.769252E+06  
 BETA      CL      CD      CM      CROLL      CN      CY      CNTR      CMTR      CLTR      CDTR  
 -0 01      0 97      -0.16      -0.30      0 00      -0 00      0 01      0.81      -0 27      0.78      0.23

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
%X/C	BP=3' 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	----TOP----	NO.	CP	
	CP	CP		CP	CP	CP	CP	CP	XLOC			
0 0	-0 2558	-0 2416	0.0	-2 1445	-3 5683	-5.7835	-2 7773	-1 4722	38.2	-0	1141	1 -0.6010
2 5	-0 2558	-0 2558	2 5	-0 2263	-1 8083	-2 4609	-2 5202	-1.2151	43 4	-0	1291	2 0 1422
5 0	-0 2558	-0 2558	5 0	-1 0371	-1 2941	-1.6304	-2 4807	-1 2348	50.4	-0	1442	3 -1.0584
10 0	-0 2558	-0 2558	10 0	-0 2460	-0 9778	-1.2151	-2 1643	-1.2348	56.1	-0	1743	4 0 0850
15 0	-0 2416	-0 2558	15 0	-0 6811	-0 7997	-1.0173	-1.6304	-1.1557	62 5	-0	2194	5 -0 9441
25 0	-0 2416	-0 2558	24 0	-0 5229	-0 6416	-0 8196	-0.8789	-1.1162	69 7	-0	2646	6 -0 0865
35 0	-0 2558	-0 2416	33.0	-0 4867	-0.4867	-0 6010	-0 6010	-1.0584	76.9	-0	2495	7 -0.2989
50 0	-0 2416		54.0	-0 3723	-0 4295	-0 4867	1 1142	-0.8297	83.4	-0	2043	8 -0.2867
56 0	-0 2558	-0 2558	65 0	-0 4603	-0 7764	-0.5054	-0.4151	-0.7011	89.4	-0	2796	9 -0.2744
65 0	-0 2558	-0 2416	78.5	1 4364	-1 0322	-0 5204	-0 5054	-0 7463	95.4	-0	2344	10 -0.2744
76 0	-0 2558	-0 2558	79.5	56 2874	58 8117	7.3344	7 3545	91 7658	101.1	-0	1893	11 -0.2621
79 0	-0 2698	-0 2558	80 5	-36.5782	-43.6660	-0 4157	-0 3342	-1 3832	---	BOTTOM---		12 -0.2867
80 5	-0 2629	-0 2324	81 3	24 6975	81 3078	-0.3546	-0 2528	-0.6398	38.2	-0	0237	
81.0	-0 2426	-0 2120	82 0	7 8228	-1 0573	-0.3750	-0 3037	-0 6295	43 4	0	0515	
82 0	-0.2426	-0.2426	84 0	2 5172	-2 1572	-0 3648	-0.3444	-0 6907	50 4	-0	0689	
84 0	-0 2528		87.0	13.8821	-2.0248	-0 4565	-0.3546	-0 2324	56.1	0	0666	
87 0	-0 2426		89 0	-6.8051	-1 1199	-0 4460	-0 4337	-0 7033	62.5	0	0967	
89 0	-0 2495	-0 2646	93 0	0 6323	-0 6910	-0 3357	-0 2621	-0.2744	69 7	0	1569	
93 0	-0 2646		96 0	-0.1274	-0 8135	-0 2621	-0.2989	-0.6297	76 9	0	1268	
96 0	-0 2646	-0.2646	100 0	-1 9285	-1 3894	-0 1642	-0 2009	-0.2499	83.4	0	0817	
100 0	-0 2646	-0 2646	96.0	0 0441	0 0932	0.1299	0.1177	-0.0294	89.4	0	0666	
96.0	-0 2646	-0 2646	84 0	0 2034	0 1544	-0 2744	0 1912	1.2694	95.4	0	0365	
84 0	-0.2646	-0 2646	73 5	-0 0865	-0 2580	0 1422	0.0850	0 0850	101.1	-0	0990	
73 0	-0 2646	-0 2646	54 0	0 0850	0 0850	0 0850	0.0850	-0.0865				
50 0	-0 2646	-0 2495	33 0	0 1994	-0 2580	0.1422	0.1422	0.0279				
35 0	-0 2558	-0 2416	24 0	0 2286	0 2088	0 2286	0 2879	0 1297				
25 0	-0 2416	-0 2558	10.0	0 4065	0 3868	0.4065	0.4659	0 3137				
10 0	-0 2416	-0 2558	5.0	0 5450	0 5252	0.5252	0 5845	0 4281				
5 0	-0 2416	-0 2698	2 5	0 6241	0 5845	0.5648	0 5845	0.4853				
2.5	-0 2698	-0 2416										

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TABLE A34 RUN 266, BW6V, DELF=0. CMU=1.0, (BN/B)W=0.25

RUN SEQ	266 1	CLAERO =	0.1704	CDAERO =	0 1889	DCLC =	0.0	DWLC =	0 2745	BASEPR =	8.1904	PAGE	630
ALPHA	0.01	PTOT	2139 41	PSTAT	2131.39	Q	8 02	VEL	82 22	YAW	0.01	H/C	2.66
BETA	-0.01	CL	0.44	CD	-0.84	CM	-0.25	CROLL	0 00	CN	-0 00	CY	0.00
												CNTR	0.23
												CMTR	-0 20
												CLTR	0.23
												CDTR	0.06

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
	BP=3 975	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----				
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP	
0.0	-0.2269	-0 2575	0.0	0.5252	0.6110	0.4394	0.4394	0.4394	38.2	-0.1456	1	-0.3670	
2 5	-0 2269	-0.2575	2 5	-0 2898	-0 1182	-0.2898	-0.3326	-0.2468	43.4	-0.0803	2	-0 2430	
5.0	-0.2269	-0.2575	5.0	-0 2039	-0.1611	-0.2468	-0.2898	-0.3326	50.4	-0.0803	3	-0.2430	
10.0	-0 2269	-0.2575	10.0	-0.2039	-0.2468	-0 3326	-0.3326	-0 2898	56.1	-0.0803	4	-0.2430	
15 0	-0 2269	-0.2575	15.0	-0 3326	-0.2898	-0.3755	-0.2898	-0 3326	62.5	-0.0803	5	-0.2430	
25 0	-0 2269	-0 2269	24.0	-0 2468	-0.2898	-0.2468	-0 2468	-0 3326	69.7	-0.1130	6	-0 2430	
35.0	-0.2269	-0.2881	33.0	-0.2430	-0.3670	-0 2430	-0.3670	-0.3670	76.9	-0.1130	7	-0.3051	
50.0	-0.2575		54.0	-0 3670	-0.3670	-0.3670	1.6169	-0.2430	83.4	-0.1456	8	-0.2785	
56.0	-0.2269	-0.2881	65.0	-0 3741	-1.0923	-0.4068	-0.3088	-0.3088	89.4	-0.2109	9	-0.2520	
65.0	-0.2575	-0 2575	78.5	2 3027	-1.3209	-0 5047	-0.4721	-0.3741	95.4	-0.2436	10	-0.2785	
76 0	-0 2575	-0 2575	79.5	115.1164	120.8364	15.8828	16.1706	197.7843	101.1	-0.1783	11	-0 2785	
79.0	-0.2269	-0.2575	80 5	-77.8725	-84 4100	-0.4607	-0.4387	-2.8902	---BOTTOM---			12	-0.2785
80.5	-0.2840	-0.2620	81.3	61.4463	180.7986	-0 3503	-0 3062	-0.3944	38.2	-0.0803			
81 0	-0.3062	-0 2840	82.0	-0 3283	5.4583	-0.4166	-0 3062	-0 3283	43.4	-0.0477			
82.0	-0.2840	-0.2399	84.0	-16.5832	-1 7196	-0.3503	-0.3503	-0.3283	50.4	-0.1130			
84.0	-0.2620		87 0	-11.5485	-3 8179	-0.4387	-0.3503	-0.3283	56.1	-0.0803			
87.0	-0.3283		89.0	6.8429	-1 7400	-0.4114	-0.4114	-0 3051	62.5	-0.0477			
89.0	-0.2762	-0.2762	93.0	0.9704	-1.1554	-0.3582	-0 2785	-0.2520	69.7	-0.0477			
93.0	-0.2109		96 0	0.0669	-1 5275	-0.2785	-0.2254	-0.2785	76.9	-0.0803			
96 0	-0 2762	-0.2762	100 0	-3.6798	-2.5106	-0.1722	-0.1457	-0.2785	83.4	-0.1130			
100 0	-0 2762	-0.2762	96.0	-0.1722	-0 0394	0.0669	0 0935	0.0935	89.4	-0.1130			
96 0	-0 2436	-0 2436	84 0	0 0669	0.0669	-0.2785	0.1466	1 8738	95.4	-0.0803			
84.0	-0.2762	-0 2762	73 5	-0.2430	-0 2430	0 1290	0.0050	0 1290	101.1	-0.1456			
73.0	-0.2436	-0 2436	54.0	-0.1191	-0 2430	-0.2430	-0.1191	-0.2430					
50.0	-0 2762	-0.2436	33.0	-0 1191	-0.2430	-0.2430	-0.2430	-0 2430					
35.0	-0.2269	-0.2269	24 0	-0 1182	-0.1611	-0.1611	-0.0753	-0.1611					
25 0	-0 2269	-0.2269	10.0	-0 0324	-0 1611	-0.1182	-0.0324	-0.2430					
10.0	-0.2881	-0.2575	5.0	-0.0753	-0 1611	-0 0753	-0 0324	-0.2430					
5 0	-0.2269	-0.2575	2 5	-0.1182	-0.1182	-0 0324	-0.0324	-0 1191					
2 5	-0 2575	-0 2575											

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TABLE A34 (Continued)

RUN SEQ 266 3  
 ALPHA 4 04  
 BETA -0.01  
 CLAERO = 0 3305  
 PTOT 2139 41  
 CL 0.68  
 CDAERO = 0 2044  
 VEL 7 91 81 65  
 YAW 0.01 2 68  
 CROLL 0 00  
 WING / CANARD  
 DCLC = 0 0  
 H/C 0 0  
 CN 0 00  
 PRESSURES  
 DWLC = 0.3465  
 CMUC 1.062  
 CMUT 1 062  
 CNTR 0 40  
 CMTR -0 23  
 PAGE 632  
 BASEPR = 8.1904  
 HGT 87 6179  
 RN 0.518690E+06  
 CLTR 0 40  
 CDTR 0 09

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3' 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP-----	NO	CP	
	CP	CP		CP	CP	CP	CP	CP				
0 0	-0 2428	-0 3049	0 0	0 2590	0 2155	-0.1760	-0 2195	-0.3935	38 2	-0.0610	1 -0 3849	
2 5	-0 2428	-0 3049	2.5	-0 2630	-0 6546	-0 8721	-0 8286	-0 9156	43 4	-0 0941	2 -0 2592	
5 0	-0.2428	-0 2738	5.0	-0 4370	-0 4805	-0 6546	-0 6111	-0 7415	50 4	-0.0610	3 -0 5107	
10 0	-0 2428	-0 2428	10 0	-0 2630	-0 4805	-0.5676	-0.4805	-0 5676	56 1	-0 0941	4 -0 1334	
15 0	-0 3049	-0 2738	15.0	-0 4370	-0 4805	-0 5241	-0.4370	-0 5241	62 5	-0.1273	5 -0 3849	
25 0	-0 2738	-0 3049	24.0	-0 3935	-0 4805	-0 4370	-0 3066	-0 4370	69.7	-0 1604	6 -0 2592	
35 0	-0 2428	-0 2428	33.0	-0 3849	-0 5107	-0 3849	-0 5107	-0 5107	76.9	-0.1935	7 -0 2683	
50 0	-0 2118		54 0	-0 3849	-0.5107	-0 5107	1.6274	-0 3849	83.4	-0.1604	8 -0 2413	
56 0	-0.2738	-0 2738	65.0	-0 4584	-1 1537	-0 4915	-0.4253	-0.4253	89.4	-0.1935	9 -0 2683	
65 0	-0 2428	-0 2738	78.5	2 3230	-1 3855	-0 5577	-0.5577	-0 4915	95.4	-0.2597	10 -0.2413	
76 0	-0.2738	-0 2738	79 5116	9989	122.7111	17.0393	18.0468	200 6927	101.1	-0 2266	11 -0 2413	
79.0	-0.2428	-0.2428	80.5-79	0231	-85.8553	-0.4576	-0.4127	-2 6979	---	---BOTTOM---	12 -0 2413	
80 5	-0 1664	-0.2111	81.3 62	3592	161 9876	-0 2784	-0.2111	-0 3680	38 2	-0.0941		
81.0	-0.1664	-0 1664	82 0	0 1921	5 6582	-0 3456	-0 2336	-0 2111	43.4	-0.0279		
82.0	-0.2111	-0 2336	84 0-16	7891	-1 6001	-0 2560	-0.2560	-0 3007	50.4	-0.0941		
84.0	-0.1440		87.0-11	.5245	-3.7732	-0 4352	-0.3007	-0 2111	56.1	-0.0279		
87 0	-0 1888		89.0	7 1169	-1 8584	-0 4569	-0.4299	-0.3490	62.5	0 0052		
89.0	-0 2928	-0 2266	93 0	0.9986	-1 2655	-0 3221	-0.2413	-0 2413	69.7	0.0052		
93 0	-0 2597		96 0	0.1091	-1.5350	-0 2683	-0.2413	-0 3221	76.9	-0.0279		
96 0	-0 2597	-0 2266	100 0	-3 7182	-2 5592	-0 1604	-0 1335	-0.2413	83.4	-0.0279		
100 0	-0 2597	-0 2597	96 0	-0.1335	-0 0256	0.1630	0 1361	0.1361	89 4	-0.0279		
96 0	-0 2597	-0 2597	84 0	0 0822	0 0552	-0 2413	0 2169	1.8880	95 4	-0.0279		
84 0	-0 2597	-0 2597	73 5	-0 2592	-0 3849	0 1182	-0.1334	0 2439	101.1	-0.1604		
73 0	-0.2597	-0 2597	54 0	-0 2592	-0 2592	-0 2592	-0 1334	-0 2592				
50 0	-0 3259	-0.2597	33 0	-0 1334	-0 3849	-0 1334	-0.1334	-0.1334				
35 0	-0.2738	-0 2428	24.0	-0 0455	-0 0020	-0 0455	0 0415	-0 0890				
25 0	-0 2428	-0 2738	10.0	0 0415	0 0415	0 0850	0 1720	-0 0076				
10 0	-0 2738	-0 2428	5 0	0 0850	0 1285	0 1720	0.2155	0 1182				
5 0	-0 2738	-0 2428	2 5	0 0850	0.1720	0 2155	0 3025	0 2439				
2.5	-0.2738	-0.2428										

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TABLE A34. (Concluded)

RUN SEQ				PROPLULSIVE		WING / CANARD		PRESSURES			PAGE 634
266 5	CLAERO =	0.7371	CDAERO =	0 3416	DCLC =	0 0	DWLC =	0 4890	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12 06	2139 48	2131 68	7.80	81 09	0 01	2 73	0 0	1 075	1 075	89.2304	0.514612E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	1 23	-0 62	-0 37	0 00	-0 00	0 01	0.85	-0.32	0.82	0.24	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP-----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0 0	-0 2906	-0.2906	0 0	-2 0007	-3 8100	-3 2804	-2.6185	-1.4711	38.2	-0.1419	1 -0 6584	
2 5	-0 2906	-0 2906	2 5	-0 3238	-1 9124	-2.3537	-2 5744	-1 2946	43.4	-0.1419	2 0 1071	
5 0	-0 2591	-0 2591	5 0	-1 1181	-1 3829	-2 0889	-2 3096	-1 1622	50.4	-0 1755	3 -1 2964	
10.0	-0 2591	-0 2906	10 0	-0 3238	-0 9857	-1 3829	-2 2214	-1.2063	56.1	-0.2091	4 -0 0205	
15 0	-0.2591	-0 2906	15 0	-0 8091	-0 8533	-1 1622	-1 9124	-1.2063	62.5	-0.2762	5 -1.0412	
25 0	-0 2906	-0 2906	24 0	-0 5886	-0 7209	-0 8533	-1.2504	-1 1622	69 7	-0.3434	6 -0.2757	
35 0	-0 3221	-0.2906	33 0	-0 5309	-0 6584	-0 7860	-0.7860	-1 0412	76.9	-0.3098	7 -0 3395	
50 0	-0.2906		54 0	-0 5309	-0 5309	-0 5309	1.5105	-0.9136	83.4	-0.2426	8 -0 3122	
56.0	-0.2906	-0 2906	65.0	-0 6122	-1 2167	-0.5450	-0.5450	-0.8137	89.4	-0.2762	9 -0.3122	
65.0	-0 2906	-0 2906	78.5	2.2093	-1 4519	-0 6458	-0.5785	-0.8473	95 4	-0.2762	10 -0.3395	
76 0	-0 3221	-0 2906	79.5	119.3748	125 2603	17.5206	18.7260	203.5313	101.1	-0.2426	11 -0.3122	
79 0	-0 2906	-0.2906	80.5	-80 2880	-88.5136	-0.4997	-0 4543	-2 7267	---BOTTOM---		12 -0.3395	
80 5	-0.1816	-0.2498	81.3	63 0174	147 8048	-0 3861	-0.2952	-0.6815	38.2	-0.0411		
81 0	-0 2724	-0.2952	82.0	1 2501	5 6360	-0.3407	-0 2952	-0.7724	43.4	-0.0075		
82.0	-0.2724	-0 2270	84 0	-15 8396	-1 9995	-0.3407	-0.3179	-0.7042	50.4	-0.1083		
84 0	-0 2498		87.0	-9 1353	-3 8403	-0.4543	-0.3861	-0.3179	56.1	-0.0075		
87 0	-0.2270		89 0	6.1951	-2 0894	-0.5035	-0.4489	-0 7223	62.5	0 0597		
89 0	-0 3434	-0.3098	93.0	0 8908	-1 2419	-0.3669	-0 3122	-0 2849	69.7	0 0932		
93 0	-0 3098		96 0	0 0706	-1 5425	-0.3122	-0 3122	-0 7771	76.9	0 0597		
96.0	-0.2762	-0.3098	100 0	-3.8119	-2 6362	-0.2029	-0.2302	-0.3122	83.4	0 0260		
100 0	-0.3098	-0.3098	96.0	-0 0935	-0 0115	0.1526	0.0979	-0 0661	89.4	-0.0075		
96 0	-0.3098	-0.3098	84 0	0 1526	0.0979	-0.3395	0.1253	1.6290	95.4	-0.0411		
84 0	-0.3098	-0 3098	73.5	-0.1481	-0.2757	0 1071	-0.0205	0.2346	101.1	-0.1755		
73 0	-0.3098	-0 3098	54 0	-0 0205	-0 1481	-0 1481	-0 1481	-0 1481				
50 0	-0 3098	-0.3434	33 0	0 1071	-0.4033	0.1071	0.1071	-0.1481				
35 0	-0 2906	-0 2906	24.0	0 1175	0 2058	0.2058	0 2941	0.0293				
25 0	-0 2906	-0 2906	10.0	0 2941	0 3382	0.3823	0 4706	0.2346				
10 0	-0 2906	-0 3221	5 0	0 4706	0 5147	0.5589	0.6030	0.3622				
5 0	-0.2591	-0 2906	2 5	0 6030	0 5147	0.5589	0.6030	0.3622				
2 5	-0 2591	-0 2906										

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TABLE A35. RUN 267, BW6V, DELF=0, CMU=2 O, (BN/B)W=0 25

RUN SEQ	267 1	CLAERO =	0 1452	CDAERO =	0.2717	DCLC =	0.0	DWLC =	0 5354	BASEPR =	8 1904	PAGE	635		
ALPHA	0 01	PTOT	2139 69	PSTAT	2135 62	Q VEL	4.07 58 56	YAW H/C	0 01 2 66	CMUC	0.0	CMUT	2.067 2.067	HGT RN	86.9890 0.371625E+06
BETA	0 01	CL	0 68	CD	-1.72	CM	-0.35	CROLL	0 01 0 00	CN	-0.00	CY	0.26	CMTR	-0 25
														CLTR	0 26
														CDTR	0.04

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **	
%X/C	BP=3 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP	
	CP	CP		CP	CP	CP	CP	CP	XLOC			
0 0	-0.3198	-0 3198	0.0	0.6692	0 6692	0.5846	0 5001	0.6692	38 2	-0.0869	1 -0.4596	
2 5	-0.2595	-0 2595	2 5	-0.2612	-0 0920	-0.3457	-0.2612	-0 0074	43 4	-0.0869	2 -0.2151	
5 0	-0 2595	-0 1991	5 0	-0 0920	-0.0920	-0 2612	-0 1766	-0 1766	50.4	-0.0869	3 -0.4596	
10 0	-0 2595	-0 2595	10.0	-0 2612	-0 0920	-0 2612	-0.0920	-0 1766	56.1	-0.0869	4 -0.2151	
15 0	-0 1991	-0 2595	15 0	-0 1766	-0 1766	-0 3457	-0 0920	-0.2612	62.5	-0.0225	5 -0.2151	
25 0	-0 3198	-0 3198	24 0	-0 1766	-0 1766	-0 2612	-0 0074	-0 2612	69.7	-0.0869	6 -0.2151	
35 0	-0 3198	-0.3198	33.0	-0.4596	-0 4596	-0 4596	-0.4596	-0.4596	76.9	-0 1512	7 -0.3549	
50 0	-0.3198		54.0	-0 4596	-0.4596	-0 4596	2 4746	-0 4596	83 4	-0.1512	8 -0.3025	
56 0	-0 2595	-0 1991	65 0	-0 5375	-1 8893	-0 4731	-0.4088	-0 3444	89.4	-0.2156	9 -0.3025	
65.0	-0 2595	-0 3802	78 5	3.7112	-1.9537	-0 6019	-0 5375	-0.4088	95 4	-0.2800	10 -0 3025	
76 0	-0.2595	-0 2595	79 5228.0465	239 2819	34 2393	37 2004	389.1824	101 1	-0.2156	11 -0 3025		
79 0	-0 2595	-0.3198	80 5*****	-166 5815	-0 4733	-0.6040	-5 0899	---	BOTTOM---	12 -0.3025		
80.5	-0.2120	-0 1249	81 3122 6918	312 1912	-0.2120	-0.1684	-0 2120	38.2	-0 0869			
81 0	-0.2555	-0 2991	82.0	-1.2572	12 4618	-0 2555	-0.1684	-0 1249	43 4	-0.0869		
82 0	-0 1249	-0 2555	84.0	-35.7949	-1 9977	-0.2991	-0 2120	-0.1249	50 4	-0.1512		
84 0	-0 2120		87 0	-26.5624	-6 8320	-0 4733	-0 2555	-0 1684	56.1	-0.0869		
87.0	-0 2120		89.0	13 1118	-3 3416	-0 5121	-0.4597	-0.3025	62.5	-0.0869		
89 0	-0 2800	-0 2800	93 0	2.1603	-2.1366	-0 3549	-0.3025	-0 3025	69.7	-0.0869		
93 0	-0 2156		96.0	0 5359	-2.7651	-0.3025	-0 2501	-0 3025	76 9	-0.0869		
96 0	-0 3444	-0 2800	100 0	-6.4857	-4.5468	-0 1453	-0 0929	-0 2501	83 4	-0 1512		
100 0	-0 3444	-0 2800	96.0	-0 3025	-0.1453	0.1167	0.1167	0.0643	89.4	-0.0869		
96 0	-0.2800	-0 2800	84.0	0 0119	0.0119	-0.3025	0.1691	2 5271	95.4	-0.1512		
84 0	-0 3444	-0.2800	73.5	-0 2151	-0 4596	0 0295	0 0295	0 5184	101.1	-0.2156		
73 0	-0 2800	-0.2800	54.0	-0.2151	-0.2151	-0.2151	-0 2151	-0.2151				
50.0	-0 2800	-0.2800	33.0	-0.2151	-0 4596	-0.2151	-0.2151	-0.2151				
35 0	-0.3198	-0 3198	24 0	0 0772	-0 0920	-0.0920	0.0772	-0.0920				
25 0	-0 3198	-0 3198	10 0	-0.0074	-0 0920	0 0772	0 1617	-0.2151				
10 0	-0 2595	-0 2595	5 0	0.2463	0 0772	-0 0074	0 1617	-0.2151				
5 0	-0.2595	-0.2595	2 5	-0.0074	-0 0074	0.2463	0.1617	0.0295				
2 5	-0.2595	-0.3198										

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TABLE A35. (Continued)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 637				
267 3	CLAERO =	0.3148	CDAERO =	0.2856	DCLC =	0.0	DWLC =	0.6698	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.03	2139.69	2135.62	4.07	58.56	0.01	2.67	0.0	2.054	2.054	87.4372	0.371525E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	0.98	-1.66	-0.39	0.01	-0.00	0.00	0.44	-0.29	0.44	0.07	

***** CANARD *****			***** WING *****						**FUSELAGE**		***MISC***	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO	CP
0.0	-0.3199	-0.3199	0.0	0.3309	0.0772	-0.7685	-0.3457	-0.5994	38.2	-0.1512	1	-0.4596
2.5	-0.2595	-0.3802	2.5	-0.2612	-0.5148	-1.0223	-0.9377	-1.1915	43.4	-0.0869	2	0.0295
5.0	-0.2595	-0.3802	5.0	-0.5148	-0.4303	-0.5994	-0.6840	-0.8531	50.4	-0.1512	3	-0.2151
10.0	-0.2595	-0.3802	10.0	-0.3457	-0.4303	-0.6840	-0.4303	-0.5994	56.1	-0.1512	4	0.0295
15.0	-0.3199	-0.3802	15.0	-0.4303	-0.4303	-0.4303	-0.3457	-0.5148	62.5	-0.1512	5	-0.2151
25.0	-0.3199	-0.3802	24.0	-0.2612	-0.4303	-0.4303	-0.2612	-0.4303	69.7	-0.1512	6	0.0295
35.0	-0.3199	-0.2595	33.0	-0.2151	-0.2151	-0.2151	-0.2151	-0.2151	76.9	-0.2156	7	-0.2501
50.0	-0.3199		54.0	-0.2151	-0.2151	-0.2151	2.7191	-0.2151	83.4	-0.1512	8	-0.2501
56.0	-0.3199	-0.3199	65.0	-0.5375	-2.0181	-0.4731	-0.4731	-0.4088	89.4	-0.2156	9	-0.3025
65.0	-0.2595	-0.3802	78.5	3.3249	-1.8893	-0.6663	-0.6019	-0.5375	95.4	-0.2800	10	-0.3549
76.0	-0.3802	-0.3199	79.5228.2211	239.4146	34.9789	38.8985	389.2314	101.1	-0.2800	11	-0.3025	
79.0	-0.3802	-0.3199	80.5*****	-166.1465	-0.5169	-0.5604	-4.9156	---	BOTTOM---	12	-0.3025	
80.5	-0.2991	-0.2120	81.3122.7361	280.3921	-0.1684	-0.1249	-0.2555	38.2	-0.0869			
81.0	-0.2120	-0.2555	82.0	-0.3862	13.0280	-0.2120	-0.1684	-0.2120	43.4	-0.0225		
82.0	-0.0814	-0.2555	84.0	-33.9641	-2.0412	-0.0378	-0.2555	-0.2555	50.4	-0.1512		
84.0	-0.2120		87.0	-25.3410	-7.0497	-0.5169	0.0929	-0.2991	56.1	-0.0225		
87.0	-0.1249		89.0	13.8453	-3.5513	-0.5121	-0.4073	-0.3549	62.5	-0.0869		
89.0	-0.3444	-0.2800	93.0	2.2127	-2.1366	-0.4073	-0.2501	-0.3549	69.7	-0.0225		
93.0	-0.2800		96.0	0.5883	-2.9225	-0.3549	-0.2501	-0.3549	76.9	-0.0225		
96.0	-0.2800	-0.2800	100.0	-6.5380	-4.5468	-0.1977	-0.1453	-0.3025	83.4	-0.0869		
100.0	-0.3444	-0.2800	96.0	-0.2501	-0.1453	0.0643	0.0643	0.1167	89.4	-0.0869		
96.0	-0.3444	-0.3444	84.0	0.0119	0.0119	-0.3025	0.1691	2.3698	95.4	-0.1512		
84.0	-0.3444	-0.2800	73.5	0.0295	-0.2151	0.2740	0.0295	1.0075	101.1	-0.2156		
73.0	-0.3444	-0.3444	54.0	0.0295	0.0295	0.0295	0.0295	0.0295				
50.0	-0.2800	-0.2800	33.0	0.0295	-0.2151	0.0295	0.2740	0.0295				
35.0	-0.3199	-0.3199	24.0	-0.0074	0.0772	-0.0074	0.2463	-0.0920				
25.0	-0.3199	-0.3199	10.0	0.1617	0.0772	-0.0074	0.3309	0.2740				
10.0	-0.3199	-0.4405	5.0	0.2463	0.2463	0.3309	0.4155	0.2740				
5.0	-0.3199	-0.3199	2.5	0.3309	0.3309	0.3309	0.4155	0.2740				
2.5	-0.3199	-0.3199										

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TABLE A35 (Concluded)

RUN SEQ 267 5  
 ALPHA 12 00  
 BETA -0 01  
 CLAEAD = 0 6999  
 PTOT 2139 62  
 CL 1 67  
 PSTAT 2135 67  
 CD -1 45  
 CDAERD = 0 4505  
 VEL 3.96 57 76  
 CM 0 48  
 WING / CANARD DCLC = 0.0  
 YAW H/C 0.01 2 72  
 CN -0 00  
 PRESSURES DWLC = 0 9673  
 CMUC 0 0  
 CMUW 2 130  
 CMUT 2.130  
 CNTR 0 90  
 CMTR -0 38  
 BASEPR = 8 1904  
 HGT 89 0625  
 RN 0.366065E+06  
 CLTR 0 87  
 COTR 0.24  
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***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
%X/C	CP	BP=3.375	%X/C	CP	BP = 2	BP = 6	BP =12	BP =16	BP = 22	XLOC	CP	NO	CP
0 0	-0 4416	-0 3795	0 0	-2 4069	-3 4508	-3.1028	-2 0591	-1.4500	38 2	-0.2061	1	-0 5233	
2 5	-0 3795	-0 3795	2 5	-0 4931	-2 4069	-2 6680	-1 7110	-1 1021	43 4	-0.2061	2	0 2314	
5 0	-0 4416	-0 3795	5 0	-1 2760	-1 4500	-2 7550	-1 7110	-1 2760	50 4	-0.2061	3	-1 2777	
10 0	-0 3174	-0 3795	10 0	-0 4931	-1 1890	-2.4939	-1.5370	-1 1890	56 1	-0.2061	4	0 2314	
15 0	-0 4416	-0 4416	15 0	-0 7541	-0 8410	-2 1460	-1 5370	-1 2760	62 5	-0.2723	5	-1 0262	
25 0	-0 4416	-0 3795	24 0	-0 7541	-0 7541	-0.9280	-1.4500	-1.2760	69.7	-0 3385	6	-0 0202	
35.0	-0 5036	-0 4416	33.0	-0 5233	-0 5233	-0.7747	-1.0262	-1 0262	76 9	-0 2723	7	-0 4694	
50 0	-0.5036		54 0	-0 5233	-0.5233	-0.2717	2 2434	-0 7747	83 4	-0.2723	8	-0 4694	
56 0	-0 3795	-0 4416	65 0	-0 7358	-2 0601	-0 6034	-0.6696	-0.8020	89 4	-0 3385	9	-0 4694	
65.0	-0 4416	-0 5036	78 5	2 9723	-2 0601	-0 6696	-0 5371	-0 8020	95.4	-0 3385	10	-0 4694	
76 0	-0 4416	-0 4416	79.5234	0284 247 0211	36 1972	40 0951	400 1919	101.1	-0.2061	11	-0.4155		
79 0	-0 4416	-0 5036	80.5*****	-172 3844	-0 6717	-0 7613	-5 0620	---	BOTTOM---	12	-0.4155		
80 5	-0 4029	-0 3133	81 3126	0636 270.3923	-0 5821	-0.4029	-0.7165	38.2	-0.0736				
81 0	-0 2238	-0 3133	82 0	-0 9853 13.2605	-0 4477	-0.4029	-0 8061	43.4	-0.0074				
82 0	-0 3581	-0 4029	84 0	-35 2999 -1 6574	-0 4477	-0 4477	-0 8957	50 4	-0 2061				
84 0	-0 3581		87 0	-25 1308 -7 4363	-0 5821	-0 3133	-0 2238	56.1	-0.0074				
87 0	-0 3133		89 0	14 1371 -3.8649	-0 5772	-0.5233	-0 9006	62.5	-0.0074				
89 0	-0 4047	-0 3385	93 0	2 0099 -2 5176	-0 4694	-0 4155	-0 4694	69.7	0 0588				
93 0	-0 4047		96 0	0 3930 -3 1104	-0 4694	-0 4155	-0 8467	76 9	-0 0074				
96 0	-0 3385	-0 4047	100 0	-6 7756 -4.7812	-0 3616	-0 4155	-0 4694	83 4	-0.0074				
100 0	-0 4047	-0 4047	96 0	-0 3616 -0 1461	0 0157	-0 0382	-0 1461	89.4	-0 0736				
96 0	-0 3385	-0 4709	84 0	0 0157 0 0695	-0 4694	-0 0382	1.7405	95 4	-0 2061				
84 0	-0 4047	-0 4047	73 5	-0 0202 -0 2717	0 2314	0 2314	0 7343	101 1	-0.2723				
73 0	-0 3385	-0 4709	54 0	0 2314 0 2314	0 2314	0 2314	0 2314						
50 0	-0 4047	-0 4047	33 0	0 2314 -0 2717	0 2314	0 2314	0 2314						
35 0	-0 5036	-0 3795	24 0	0.2029 0 0289	0 1159	0.2899	-0 0581						
25 0	-0 5036	-0 4416	10 0	0 3769 0 2899	0 3769	0 4639	0 4828						
10 0	-0 4416	-0 3795	5 0	0 5508 0.3769	0 4639	0.6379	0 4828						
5 0	-0 3795	-0 3795	2 5	0.7248 0.4639	0 5508	0 5508	0 4828						
2 5	-0 3174	-0 3795											

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TABLE A36. RUN 271, BC1W6V, DELF=0, CMU=0, (BN/B)W=0.25

RUN SEQ	271 2	CLAERD = 0	1509	CDAERD = 0.0576	DCLC = 0.0	DWLC = 0.0	BASEPR = 8.1904
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	HGT RN
0 11	2139 20	2109.14	30.06	160 29	0 01	2.67	87.4555 0.998841E+06
BETA	CL	CD	CM	CROLL	CN	CY	CLTR
-0 01	0 15	0.06	-0.11	0 00	-0.00	0.01	0.15 0.09

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**			
BP=3:375 BP=10.125			BP = 2		BP = 6		BP = 12		BP = 16		BP = 22		-----TOP-----	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	CP	XL0C	CP	NO.	CP
0 0	0.3561	0.4051	0.0	0.5548	0.5319	0.5090	0.5090	0.5205	38.2	-0.0967	1	-0.2234		
2.5	-0.0033	-0.0196	2 5	-0.1779	-0.1207	-0.2123	-0.1207	-0.1207	43.4	-0.0880	2	-0.1241		
5.0	-0.0686	-0.1013	5.0	-0.0978	-0.1550	-0.2237	-0.1894	-0.2008	50.4	-0.0819	3	-0.2234		
10 0	0 9441	-0 1829	10 0	-0.1779	-0.2237	-0 2352	-0.2352	-0.2008	56.1	-0.0531	4	-0.1572		
15 0	-0 2075	-0.2238	15 0	-0.1665	-0.2123	-0.2466	-0.2581	-0 2237	62 5	-0.0531	5	-0.1903		
25 0	-0.1829	-0 2156	24 0	-0.1665	-0.2237	-0 2123	-0.2008	-0.2123	69.7	-0.0880	6	-0.1572		
35.0	-0.1666	-0 1666	33.0	-0.1903	-0 1903	-0 2234	-0.2565	-0 2234	76.9	-0.0793	7	-0 2046		
50 0	-0 1585		54 0	-0 1903	-0 1903	-0 2565	0.9351	-0 2234	83 4	-0 0531	8	-0.1904		
56 0	-0.1666	-0.1503	65.0	-0 2187	-0.2536	-0 2884	-0.2710	-0.2449	89.4	-0.0880	9	-0.2046		
65.0	-0.2156	-0 1829	78.5	1.2017	-0.4801	-0 3581	-0.3756	-0.3145	95.4	-0.1664	10	-0 1904		
76.0	-0.1829	-0.2973	79 5	-0 4313	-0.4549	1.3669	7.9112	53 5396	101.1	-0.1403	11	-0 1975		
79 0	-0.4117	-0 4117	80.5	-0 4313	-0 4490	-0 2132	-0 2840	-0.9148	---BOTTOM---			12	-0 1975	
80.5	-0 4196	-0 4078	81.3	-0.4373	53.5396	-0.1896	-0.1896	-0.3016	38.2	-0.0619				
81.0	-0 4196	-0.4078	82.0	-0 3842	-0 9148	-0 2073	-0.2132	-0.2191	43.4	-0 0096				
82.0	-0.4313	-0 3842	84.0	-0 2014	-0.2604	-0 2427	-0.2899	-0.2604	50 4	-0.0793				
84 0	-0.1483		87.0	-0.2368	-0.2780	-0 3252	-0.2899	-0.1955	56.1	-0.0183				
87.0	-0 1189		89.0	-0.2613	-0.2897	-0.3535	-0.3819	-0.2755	62.5	-0.0183				
89.0	-0.1403	-0 1664	93.0	-0 2187	-0 2258	-0 2613	-0.2046	-0 1975	69.7	-0.0619				
93 0	-0.1315		96.0	-0.1762	-0.1691	-0.1833	-0 1975	-0.2471	76.9	-0.0531				
96.0	-0.0880	-0 1054	100.0	-0.1336	-0.1194	-0.0840	-0.0840	-0 1904	83.4	-0.0444				
100 0	0 8357	-0 1141	96 0	0 0650	0.1217	0 1359	0 1430	0 1288	89.4	-0.0357				
96 0	0.0776	0.1037	84 0	0 1359	0 0934	-0.2258	0.1643	1 1645	95.4	-0.0008				
84 0	0.8357	-0.1141	73 5	-0.0248	-0 1572	0 0414	0.0745	0.2069	101.1	-0.0705				
73 0	0 0776	0 1037	54.0	-0 0910	-0 0910	-0 1241	-0.1241	-0 1241						
50 0	0.1299	0.1299	33 0	-0 1241	-0 1572	-0.1572	-0.1572	-0.1572						
35.0	-0 0115	-0 0278	24.0	-0.1321	-0 1665	-0.1665	-0.0978	-0 1550						
25 0	-0 1829	-0 2156	10.0	-0.1092	-0.1321	-0 1207	-0.0978	-0 1572						
10 0	-0.1666	-0.2075	5 0	-0.0978	-0.1665	-0.1321	-0.0520	-0 1572						
5 0	-0 2238	-0 2728	2.5	-0 0863	-0.1436	-0.0863	-0.0520	-0.0910						
2 5	-0.3708	-0.3790												

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TABLE A36 (Continued)

RUN SEQ 271 4  
 ALPHA 4 02  
 BETA -0 01  
 CLAE RD = 0 3537  
 PI TOT 2139 13  
 CL 0 35  
 PSTAT 2109 07  
 CD 0.08  
 CDAER D = 0 0769  
 Q VEL 30 06 160 44  
 CM -0 07  
 WING / CANARD DCLC = 0.0  
 YAW H/C 0 01 2 66  
 CN -0 00  
 PRESSURES CMUC 0 0  
 CMUW 0 0  
 CMUT 0.0  
 CNTR 0.36  
 CMTR -0 07  
 DWLC = 0 0  
 BASEPR = 8.1904  
 HGT 86 9525  
 RN 0.996487E+06  
 CLTR 0 35  
 CDTR 0.11

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
%X/C	BP=3.375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XLOC	CP	NO.	CP
0 0	0.3610	0 1241	0.0	0.4599	0 3340	-0.0095	-0 4102	-0 5705	38 2	-0.1436	1	-0.2929
2 5	-0 5375	-0.8152	2.5	-0 2042	-0.4217	-0.7652	-0 7766	-0 9713	43.4	-0.1262	2	0 0050
5 0	-0 4477	-0 6845	5 0	-0 2385	-0 3988	-0 5019	-0 6278	-0.6965	50 4	-0.0913	3	-0 0281
10 0	0 9082	-0.5048	10 0	-0 1927	-0 3873	-0 4675	-0.4789	-0 5133	56.1	-0 0652	4	-0 0281
15 0	-0 4231	-0 4395	15 0	-0 2958	-0.3186	-0 4217	-0 4560	-0.4446	62 5	-0.0739	5	-0 2929
25 0	-0 3415	-0.3496	24.0	-0 2499	-0.3072	-0.3415	-0.3415	-0.3645	69 7	-0.1174	6	-0.0943
35 0	-0 2435	-0 2353	33 0	-0.2267	-0 2598	-0 2929	-0.2929	-0 2929	76.9	-0.1087	7	-0.2221
50 0	-0 2516		54 0	-0 1936	-0 2598	-0 2929	0 8987	-0 2929	83 4	-0 0739	8	-0 2079
56 0	-0 2190	-0 2190	65 0	-0 2569	-0.3004	-0 3353	-0.3179	-0.3179	89 4	-0.1610	9	-0 2150
65 0	-0 2598	-0 2680	78 5	1 1723	-0 4922	-0 3789	-0 3789	-0.3876	95.4	-0.1784	10	-0.2150
76 0	-0 1945	-0.3496	79 5	-0.4347	-0.4406	1 4049	8.4271	53.5613	101.1	-0.1523	11	-0.2221
79.0	-0.4395	-0 4395	80.5	-0 4465	-0 4524	-0.2224	-0.2932	-0 8651	---	BOTTOM---	12	-0.2150
80 5	-0 4347	-0 4347	81.3	-0.4583	53 5613	-0 2224	-0.2165	-0 3521	38.2	-0.0390		
81.0	-0 4288	-0 4288	82 0	-0.4052	-0 8356	-0 2460	-0 2401	-0.2696	43.4	0.0220		
82 0	-0.4111	-0 3463	84 0	-0 2224	-0.2401	-0.2519	-0 2755	-0.3050	50.4	-0.0652		
84 0	-0 1753		87 0	-0 2401	-0.2932	-0.3345	-0 2873	-0 1458	56.1	0.0046		
87 0	-0 1576		89 0	-0 2930	-0 3143	-0.3498	-0.3640	-0 3569	62 5	0 0046		
89 0	-0 1523	-0 1872	93 0	-0.2363	-0 2504	-0 2434	-0.2150	-0 2150	69.7	0 0133		
93.0	-0.1349		96 0	-0 1937	-0 1937	-0 1795	-0.2008	-0 3214	76.9	-0.0129		
96 0	-0 1000	-0.1262	100 0	-0.1653	-0 1228	-0 0802	-0.1086	-0.2079	83.4	-0 0129		
100 0	0 6930	-0 0913	96.0	0.0830	0 1326	0 1610	0.1397	0 1113	89 4	0.0046		
96 0	0 1091	0 1004	84.0	0 1468	0 1042	-0 2008	0 1894	1 1186	95 4	0 0133		
84.0	0 6930	-0 0913	73 5	0.0381	-0 1605	0 1043	0 1043	0 1705	101 1	-0.0739		
73 0	0 1091	0 1004	54 0	-0 0612	-0 0612	-0 0612	-0 0612	-0 0943				
50 0	0 1788	0 1527	33 0	-0 0281	-0 1605	-0 0281	-0 0281	-0 0612				
35 0	0 0016	-0 0556	24 0	-0 0324	-0.0668	-0 0439	-0 0095	-0.1011				
25 0	-0 1209	-0 2925	10.0	0 0134	0.0019	0 0363	0 1279	0 1043				
10 0	-0 0883	-0 0964	5 0	0 0821	0 0592	0 1393	0 1851	0 1705				
5 0	-0 1454	-0 1046	2 5	0 0706	0 0935	0 2195	0 2882	0 3360				
2 5	0 0996	0 1568										

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TABLE A36. (Concluded)

RUN SEQ 271 6  
 ALPHA 12 07  
 BETA -0.01  
 CLAERO = 0 8522  
 PTOT 2138 99  
 PSTAT 2108.93  
 CL 0 85  
 CD 0 20  
 CDAERO = 0 2007  
 VEL 30.06 160 57  
 CM 0 00  
 WING / CANARD  
 DCLC = 0 0  
 YAW H/C 0 01 2 71  
 CROLL CN -0.00  
 PRESSURES  
 DWLC = 0 0  
 CMUC 0 0  
 CMUW 0.0  
 CMUT 0.0  
 CY 0 01  
 CNTR 0 87  
 CMTR -0 00  
 PAGE 644  
 BASEPR = 8.1904  
 HGT RN  
 88.7508 0.994565E+06  
 CLTR CDTR  
 0.84 0.23

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3 375	BP=10.125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	----TOP----	NO	CP	
	CP	CP		CP	CP	CP	CP	CP	XLOC			
0 0	-4.7896	-1 7838	0.0	-0.3268	-1.5518	-3 9333	-3.7044	-1.5747	38.2	-0.2536	1 -0.5213	
2.5	-2 1759	-1 5551	2 5	-0.2352	-1.1168	-1.8152	-3.3494	-1.3916	43.4	-0.2013	2 0.1076	
5 0	-1.4653	-1.4489	5 0	-0.6130	-0 8878	-1.3343	-2.9030	-1.3916	50.4	-0.1490	3 -0.8854	
10.0	0.8707	-1.3999	10 0	-0.2123	-0 7275	-0.9908	-0 9107	-1.3114	56.1	-0.1490	4 0.0745	
15 0	-0 9344	-1.3428	15.0	-0 5557	-0 6702	-0.8535	-0.6931	-1.2427	62.5	-0.1838	5 -0.8854	
25.0	-0 6812	-1 3673	24.0	-0.4870	-0.5443	-0.6702	-0.6244	-1.0710	69.7	-0.2361	6 -0.0910	
35.0	-0 5342	-1.1467	33 0	-0 4551	-0.5213	-0.5544	-0.5544	-1 1171	76.9	-0.2187	7 -0.2826	
50 0	-0.3953		54 0	-0 3558	-0 4220	-0.4220	0.7697	-0 8854	83.4	-0.1751	8 -0 2755	
56.0	-0.3871	-0 5178	65.0	-0.3668	-0.4104	-0.4366	-0.4104	-0.8026	89.4	-0.2100	9 -0.2755	
65 0	-0.3708	-0.4607	78 5	1.1147	-0.5760	-0.4801	-0.4714	-0 7416	95.4	-0.2274	10 -0 2755	
76 0	-0 2565	-0 4280	79 5	-0 5198	-0 5257	1.3905	8.9963	53.5646	101.1	-0.1926	11 -0 2826	
79 0	-0 5015	-0 5097	80.5	-0 5198	-0 4962	-0.3193	-0 3665	-0 8912	---	---BOTTOM---	12 -0.2826	
80 5	-0 4785	-0 5552	81 3	-0.5080	52.7030	-0 2958	-0 3135	-0 6672	38.2	0.0427		
81 0	-0 4962	-0 4078	82 0	-0 4667	-0 8382	-0 3135	-0 3135	-0 7085	43.4	0.0863		
82 0	-0 4196	-0 3193	84.0	-0 2722	-0 2899	-0 3193	-0 3429	-0 7026	50 4	-0.0270		
84.0	-0.2132		87.0	-0 2899	-0.3193	-0 3665	-0.3429	-0.2309	56.1	0.0689		
87.0	-0.2191		89.0	-0 3464	-0.3677	-0.4103	-0.3890	-0.7224	62.5	0.1038		
89 0	-0.2187	-0 3146	93.0	-0 2826	-0.2826	-0.3039	-0.2755	-0 2755	69.7	0.1212		
93 0	-0.1838		96.0	-0 2400	-0.2329	-0 2259	-0.2329	-0.6018	76.9	0.1212		
96.0	-0 1490	-0 2623	100 0	-0 1904	-0 1336	-0.1123	-0.1691	-0 2613	83.4	0.0776		
100.0	0 5830	-0.2710	96.0	0 0934	0 1430	0.1501	0.1218	-0 0130	89.4	0.0602		
96.0	0.1125	0 0340	84 0	0 1927	0.1430	-0 2684	0.1714	1 0368	95.4	0.0515		
84.0	0.5830	-0 2710	73 5	0 1076	-0.2896	0.1076	0.1076	0.1738	101.1	-0.0792		
73.0	0 1125	0 0340	54 0	0 0745	0 0414	0 0414	0 0083	-0 0579				
50 0	0 2258	0 1125	33.0	0 1076	-0.2896	0 1076	0.1076	-0 0248				
35.0	0.0702	-0 0033	24.0	0 1541	0.1541	0.1885	0.2114	0.0854				
25 0	0 0376	-0 4280	10.0	0.2800	0 2800	0.3716	0.4060	0.2731				
10 0	0.1356	0 0947	5.0	0 3716	0.3831	0 4289	0.5205	0.4055				
5 0	0.1438	0.1438	2 5	0 4861	0.4403	0.4976	0 5205	0 4386				
2 5	0.5685	0 5766										

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RUN SEQ				PROPLULSIVE		WING / CANARD		PRESSURES			PAGE 645
272 1	CLAERO =	0 2138	CDAERO =	0 1794	DCLC =	0 0491	DWLC =	0 1273	BASEPR =	8 1904	
ALPHA	PTOT	PSTAT	0	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0 03	2138 77	2120 02	18 76	126 58	0 01	2 66	0.282	0.491	0 773	87.0914	0.787667E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	0 39	-0 57	-0 19	0 00	-0 00	0 01	0 22	-0 15	0.22	0 05	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3	BP=10		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO CP	
0 0	0 4001	0 5179	0 0	0 2683	0 3234	0.4334	0.4885	0.4885	38.2	-0.1568	1 -0 2708	
2 5	-0 0448	-0 1365	2 5	-0 3187	0 3050	-0 0619	-0.1903	-0.2821	43.4	-0 1429	2 -0.1117	
5 0	-0.1234	-0 1888	5 0	0.4518	0.1766	-0.1353	-0 2086	-0.2454	50.4	-0.0731	3 -0.2178	
10 0	0.8844	-0 2150	10 0	-0 3004	-0 0252	-0.2637	-0.2454	-0.2821	56.1	-0.0591	4 -0.1117	
15 0	-0 2673	-0 2542	15 0	-0 0436	-0.1536	-0 2454	-0 2454	-0.2821	62.5	-0.0452	5 -0.2178	
25.0	-0 2412	-0 2281	24 0	-0 1720	-0.2086	-0.2454	-0 1903	-0.2454	69.7	-0.0731	6 -0.1647	
35 0	-0 2281	-0 2542	33.0	-0.1647	-0 2178	-0 2178	-0.2178	-0.2178	76.9	-0 0591	7 -0.3618	
50.0	-0 2542		54.0	-0 1647	-0 2178	-0 2178	0.8961	-0 2178	83.4	-0.0731	8 -0.3618	
56.0	-0 2804	-0 2673	65.0	-0.3105	-0.5478	-0 3943	-0 3244	-0 2825	89.4	-0.1568	9 -0 3391	
65 0	-0 3851	-0 4374	78.5	1 2675	-1.3717	-0.6177	-0.4640	-0.3663	95.4	-0 2267	10 -0 3618	
76.0	-0 3066	-0.7778	79.5	61 5331	64 5663	2.1076	9 2500	85.2284	101.1	-0.2127	11 -0.3618	
79 0	-1 1835	-1.2489	80 5-	32 5369	-45.2721	-0 2260	-0.3110	-1 4447	---	BOTTOM---	12 -0.3504	
80.5	-17.6854	-25 5552	81.3	-3 3437	85.2284	-0 2543	-0.1977	-0 2733	38.2	-0 0591		
81 0	0 3220	6.4630	82.0	30 3652	-8 3698	-0.3016	-0 2449	-0.2071	43.4	-0 0172		
82.0	0 3031	-0 6890	84 0	3 0523	0.1614	-0 4055	-0.3110	-0.2638	50.4	-0.1010		
84.0	0 2086		87 0	-9.3427	-2 0589	-0 5378	-0.3205	-0 2638	56.1	-0.0172		
87 0	1 5123		89 0-	11 4670	-1 1461	-0 5892	-0.4300	-0.3277	62.5	-0.0172		
89 0	0 2341	-1 0785	93.0	3 5938	-0 8051	-0.4414	-0 3618	-0.3504	69.7	-0 0452		
93 0	0 4715		96 0	-0 3504	-0 8733	-0 3163	-0.2481	-0.2709	76.9	-0.0591		
96.0	1 1977	0 3598	100.0	-1.9304	-1.2484	-0 1572	-0.1345	-0 3618	83.4	-0.0591		
100 0	0 3878	-1 5812	96.0	-0 0663	0 0247	0 0815	0 1042	0 1042	89.4	-0 0452		
96 0	-0 0871	-0 0452	84 0	0 0701	0 0474	-0.3504	0 1270	0.9340	95.4	-0 0312		
84 0	0 3878	-1 5812	73 5	-0 2178	-0 2708	0 1535	0 1005	0.2596	101.1	-0.1289		
73 0	-0.0871	-0 0452	54 0	-0 0586	-0 1117	-0 1117	-0 1117	-0.1117				
50 0	0 0945	0 0945	33.0	-0 1117	-0 2708	-0 1647	-0 1117	-0.1117				
35 0	-0 0448	-0 0710	24.0	-0 1353	-0 1903	-0 1536	-0 0802	-0 1536				
25.0	-0 2019	-0 3851	10 0	-0 1170	-0 1720	-0.1353	-0 0252	-0.1117				
10 0	-0 1888	-0 2019	5 0	-0.1536	-0 2086	-0.1170	-0.0252	-0.1117				
5 0	-0 2673	-0 2542	2.5	-0.2637	-0 2821	-0 1536	-0 0986	-0 0056				
2 5	-0 3066	-0 3066										

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TABLE A37 (Continued)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 647				
272 3	CLAERD =	0.3997	CDAERD =	0.2141	DCLC =	0.0707	DWLC =	0.1654	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.03	2138.63	2120 21	18 42	125.45	0.01	2.66	0.291	0 507	0.799	87.0853	0 780197E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	0.64	-0.55	-0.16	0 00	-0.00	0.01	0.43	-0.11	0.42	0.08	

***** CANARD *****			***** WING *****						**FUSELAGE**		***MISC.***	
	BP=3.375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XL0C	CP	NO	CP
0.0	0.2659	0 1059	0 0	0 4944	0 4197	0.1021	-0 1408	-0.2716	38.2	-0.1814	1	-0 2975
2 5	-0 6138	-0 8404	2.5	-0.3276	0 0087	-0.5705	-0.7574	-0.9255	43.4	-0.1672	2	-0.0274
5.0	-0 4806	-0 6405	5 0	0 2142	-0.1034	-0.4584	-0 5705	-0.7387	50.4	-0.1103	3	-0 3515
10.0	1.1455	-0 6005	10.0	-0 3463	-0.3090	-0 4584	-0.4771	-0.5332	56.1	-0.0819	4	-0 0274
15 0	-0 4539	-0 5605	15 0	-0 1595	-0.3837	-0.4211	-0.4211	-0.4958	62.5	-0.0677	5	-0.2975
25 0	-0 3872	-0.4139	24 0	-0.2155	-0.3276	-0 3650	-0.3463	-0.3837	69.7	-0.0961	6	-0.1354
35.0	-0.3206	-0.3339	33.0	-0.2435	-0 2975	-0.3515	-0 2975	-0 2975	76.9	-0.0961	7	-0 3555
50.0	-0.3339		54.0	-0 2435	-0.2975	-0.3515	1.0529	-0.2975	83.4	-0.0819	8	-0.3555
56 0	-0 3473	-0.3339	65.0	-0.3663	-0.5939	-0.3948	-0.3521	-0.3521	89.4	-0.1814	9	-0.3555
65 0	-0.4139	-0.4939	78.5	1 4682	-1.4044	-0.5512	-0.4659	-0 4090	95.4	-0.2383	10	-0.3670
76.0	-0.3339	-0 8537	79.5	62 6819	65.6352	2.1439	9.7834	86.7753	101.1	-0.2099	11	-0.3323
79 0	-1.2536	-1 3202	80 5	-33.0613	-45.9539	-0.2807	-0.3866	-1 4257	---BOTTOM---			
80.5	-18.4560	-25 9891	81 3	-3 7541	82 7416	-0.3385	-0.2615	-0.3577	38.2	-0 0250	12	-0.3439
81.0	0.9412	6.7910	82 0	31.0467	-8.4300	-0 3577	-0.2711	-0.3000	43.4	0.0176		
82.0	0 6044	-0 9446	84.0	3.0290	-0.0594	-0.3866	-0.3385	-0.3289	50.4	-0.0819		
84 0	0.1522		87.0	-9 4788	-2.2243	-0 4828	-0.3192	-0 3096	56.1	0.0319		
87.0	1 4992		89 0	-11.8040	-1.2121	-0.5175	-0.4018	-0.3786	62 5	0.0319		
89.0	0.2452	-1.2338	93.0	3.7540	-0 7838	-0.3670	-0.3439	-0.3439	69.7	-0.0108		
93 0	0.4727		96.0	-0 0198	-0 8879	-0.2629	-0.2397	-0 3323	76.9	-0.0108		
96.0	1.1980	0 2736	100.0	-1.9298	-1.2468	-0.1239	-0.1355	-0 3439	83.4	-0.0250		
100 0	0.6292	-1.6178	96 0	-0 0429	0 0381	0.1423	0.1307	0.0960	89 4	-0.0108		
96.0	-0.0392	-0 0108	84.0	0 1075	0 0844	-0.3439	0.1770	1 1494	95.4	-0.0108		
84 0	0.6292	-1.6178	73.5	-0 1895	-0.2975	0.1887	0.0806	0.1346	101.1	-0.1245		
73 0	-0.0392	-0 0108	54.0	-0 0274	-0.0814	-0.0814	-0.0814	-0.1354				
50 0	0 1457	0.1314	33 0	-0.0814	-0.2975	-0 0814	-0.0274	-0 0814				
35.0	-0 0007	-0.0674	24 0	-0.1034	-0.1034	-0 0847	-0 0287	-0 0847				
25 0	-0.1207	-0 4406	10.0	-0 0847	-0 0847	-0.0100	0.0647	0 0266				
10.0	-0.0940	-0.1074	5 0	-0.0661	-0.0661	0 0647	0.1768	0.1887				
5.0	-0.1607	-0 1207	2 5	-0.1221	-0.1221	0.1208	0.2142	0.2427				
2 5	0.0793	0.1859										

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TABLE A37 (Concluded)

RUN SEQ		PROPLULSIVE WING / CANARD PRESSURES										PAGE 649	
272	5	CLAERO =	0 9372	CDAERO =	0 3358	DCLC =	0 1093	DWLC =	0 2310	BASEPR =	8 1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
12 14	2138 35	2120 38	17.97	123 92	0.01	2 73	0 290	0 506	0 797	89.2004	0.770235E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
-0 01	1 28	-0 38	-0 08	-0 00	-0.01	0 02	0 99	-0.03	0 97	0 22			
***** CANARD *****													
	BP=3 375	BP=10.125	***** WING *****										***** FUSELAGE**
%X/C	CP	CP	%X/C	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22		----	TOP----	**MISC.**	
				CP	CP	CP	CP	CP		XLOC	CP	NO.	CP
0 0	-5.3245	-2 3869	0 0	0 4271	-0.2816	-3.1546	-4 3230	-1.6606		38.2	-0.3541	1	-0 6041
2 5	-2 2776	-1 9770	2 5	-0 3773	-0.7412	-1 6606	-3 5760	-1 4499		43 4	-0.2958	2	0 1158
5 0	-1.6900	-1.9633	5 0	-0 3007	-0 6646	-1.1626	-2.5034	-1 3925		50 4	-0.2083	3	-0 6041
10 0	1 4800	-1 8813	10.0	-0 3773	-0 6838	-0.9902	-0.8753	-1.5266		56.1	-0.1645	4	0.1158
15 0	-1 0615	-1.7447	15.0	-0 5306	-0 6838	-0.8178	-0 8178	-1 3733		62.5	-0.1937	5	-0.9364
25 0	-0 8019	-1 7993	24 0	-0.5497	-0 6071	-0.6838	-0 6455	-1.2392		69 7	-0 2228	6	-0 1058
35 0	-0 6515	-1 4031	33.0	-0 5488	-0 6041	-0.6041	-0.6595	-1.1025		76 9	-0 2228	7	-0.4579
50 0	-0.5696		54 0	-0 4380	-0 4934	-0 5488	1 2233	-0.8810		83.4	-0.2083	8	-0 4342
56 0	-0 5559	-0.6106	65 0	-0 5144	-0 7623	-0 5290	-0 5144	-0.8206		89.4	-0.2666	9	-0 4342
65 0	-0.6106	-0 6242	78 5	1 4683	-1.4766	-0.6019	-0.5873	-0.9081		95.4	-0 2958	10	-0.4460
76 0	-0 3920	-0 8292	79.5	64 5035	67 4644	2 3136	10 9539	88.9470		101.1	-0.2520	11	-0 4342
79 0	-1 3894	-1 2391	80.5	-33 8858	-47.2312	-0 4482	-0.5074	-1.5036		----	BOTTOM----	12	-0.4342
80 5	-19 3366	-26 5767	81 3	-4 3738	76 9089	-0 4186	-0 3988	-0 7244		38.2	0 0250		
81 0	1 1596	7 2061	82 0	32 0125	-8 8518	-0.4580	-0.4186	-0 6947		43.4	0 0687		
82 0	0 5875	-0 9710	84 0	3.3197	0 2324	-0 4679	-0.4778	-0 7835		50.4	-0 0771		
84 0	0.0746		87 0	-10 6868	-2.3025	-0 5074	-0.4383	-0 3988		56.1	0.0687		
87 0	1 4555		89 0	-12 0876	-1 2767	-0 5410	-0 5172	-0.7190		62 5	0 0833		
89 0	0 1416	-1.2726	93 0	3.5057	-0 8139	-0.4223	-0.4342	-0 4223		69 7	0.0833		
93 0	0 3749		96 0	-0 3867	-0 8970	-0 3036	-0.3155	-0 6715		76.9	0.0687		
96 0	1 1184	0 2728	100.0	-1 9887	-1 3123	-0.1612	-0 2205	-0.4223		83 4	0.0541		
100.0	0.8560	-1 6808	96.0	-0 0069	0 0880	0.1592	0 0999	-0 0188		89 4	0.0396		
96 0	-0 0042	-0 0333	84 0	0 1711	0 1236	-0 4342	0.1711	1 2154		95 4	0 0250		
84 0	0.8560	-1 6808	73 5	-0 1058	-0 4380	0 1711	0 1158	0 2819		101.1	-0.1354		
73 0	-0.0042	-0 0333	54.0	0 0050	0 0050	0 0050	0 0050	-0.1058					
50 0	0 1999	0.1271	33 0	0 1158	-0 3826	0 1158	0 1158	0 0050					
35 0	0 0863	-0 0094	24 0	0 1207	0 1207	0 1590	0 2548	0 1207					
25 0	0 0316	-0 6379	10 0	0 1973	0 1973	0 3122	0 4271	0 2819					
10 0	0 1546	0 0999	5 0	0 2165	0 2548	0.4271	0 5229	0 4480					
5 0	0 1409	0.1409	2.5	0.2548	0 3122	0.4655	0 5038	0 4480					
2 5	0 5782	0 5645											

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TABLE A38. RUN 273, BC1W6V, DELF=0, CMU=1.0, (BN/B)W=0 25

RUN SEQ			PROPLULSIVE WING / CANARD PRESSURES							PAGE 650	
273 1	CLAERO =	0 2216	CDAERO =	0 3187	DCLC =	0 1037	DWLC =	0.2683	BASEPR =	8.1904	
	ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	
0 06	2138 35	2129 31	9.04	87 76	0.01	2 66	0 594	1.033	1.626	87.0442 0.547491E+06	
	BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR CDTR	
-0 01	0 59	-1 26	-0.30	0 00	-0 00	0.02	0.24	-0.20	0 24	0.05	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **	
	BP=3 375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%λ/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLLOC	CP	NO	CP
0 0	0 3819	0.4090	0 0	-0 4780	-0 1736	0.3593	0 4735	0.4735	38.2	-0.1814	1	-0 3141
2 5	0 0017	-0 2155	2 5	-0 4780	1.1586	0 0929	-0 1355	-0 2116	43.4	-0.2104	2	-0.0940
5 0	-0 1341	-0.2427	5.0	1 4631	0.7400	-0.1736	-0.2116	-0.3258	50.4	-0.1234	3	-0 3141
10 0	2.5541	-0.2698	10.0	-0 5161	0.0168	-0 3258	-0.2496	-0 3639	56.1	-0.0945	4	-0 0940
15 0	-0 3241	-0 2970	15 0	-0.0974	-0.2116	-0.3639	-0.2116	-0 3639	62.5	-0.0366	5	-0.2041
25 0	-0.3241	-0 2970	24 0	-0 2877	-0 3258	-0 4780	-0 2116	-0 3639	69 7	-0.0945	6	-0.0940
35 0	-0.3241	-0 3241	33.0	-0.2041	-0 2041	-0.3141	-0 3141	-0.3141	76.9	-0.0945	7	-0.5029
50 0	-0.3784		54.0	-0.3141	-0.3141	-0.3141	1.6666	-0.3141	83.4	-0.0945	8	-0 5029
56.0	-0.4327	-0.3784	65.0	-0.4132	-0.9637	-0.5291	-0.4132	-0.2973	89.4	-0.1814	9	-0 4792
65 0	-0.5413	-0 5956	78 5	2.0493	-2.2963	-0.9347	-0.6740	-0 3842	95.4	-0.2973	10	-0.5029
76 0	-0.5142	-1.1930	79 5	130.0376	135 0918	4.8677	24.4680	175.6912	101.1	-0.2683	11	-0.5029
79 0	-1.8719	-1.9803	80 5	-66 9686	-91 2729	-0 2677	-0 5813	-2.6785	---	---	12	-0.5029
80 5	-37.7052	-52 7184	81.3	-8 8723	165.7266	-0.2872	-0.2284	-0.3068	38.2	-0.1234		
81.0	3.3193	15 0597	82 0	65 2964	-13.2427	-0.3852	-0.2677	-0.2088	43.4	-0.0656		
82 0	2 1040	-1 3261	84 0	6 7494	1.6532	-0.5421	-0 3852	-0.1892	50.4	-0.1814		
84.0	0 4968		87 0	-17 6731	-4 0897	-1.0125	-0.3656	-0.4048	56.1	-0.0366		
87 0	3 5936		89 0	-23 6839	-2 1300	-0.9037	-0.5500	-0.3614	62.5	-0.1524		
89 0	0 7746	-1 8328	93 0	8.2696	-1 1396	-0.6443	-0.5029	-0 5736	69.7	-0.1524		
93 0	1.2092		96 0	0.4640	-1.5168	-0 4321	-0.2906	-0.3614	76.9	-0.1814		
96 0	2 5998	0.9485	100 0	-3 7336	-2.5781	-0 2199	-0.1256	-0.5500	83.4	-0.0945		
100 0	0 8905	-3 0785	96 0	-0 1727	-0 0076	0 1574	0 0867	0.1103	89.4	-0.0945		
96 0	-0 2683	-0 2394	84.0	0 0867	0.0395	-0.4792	0.1339	1.3130	95.4	-0.0656		
84 0	0 8905	-3 0785	73 5	-0 3141	-0 5342	0 1261	0.0160	-0.0940	101.1	-0.2394		
73 0	-0.2683	-0 2394	54.0	-0.0940	-0 0940	-0 0940	-0.0940	-0.2041				
50 0	0 0503	0 0503	33.0	-0.0940	-0.4242	-0.0940	-0.0940	-0 0940				
35 0	-0.2427	-0 1882	24 0	-0.2496	-0.2877	-0.3258	-0 1355	-0 2877				
25 0	-0.3784	-0 5413	10 0	-0.1736	-0.2877	-0.3639	-0 1355	-0 2041				
10.0	-0.2427	-0.3241	5 0	-0.2877	-0 3639	-0 2496	-0.0974	-0.0940				
5 0	-0 2970	-0 2970	2 5	-0 5161	-0 6683	-0.3258	-0.1355	-0 2041				
2.5	-0.4327	-0 4327										

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TABLE A38. (Continued)

RUN SEQ 273 3    CLAERO = 0 4296    CDAERO = 0.3226    DCLC = 0 1441    DWLC = 0.3350    BASEPR = 8.1904    PAGE 652  
 ALPHA PTOT PSTAT Q VEL YAW H/C CMUC CMUW CMUT HGT RN  
 4.02 2138.35 2129.31 9.04 87.73 0 01 2.66 0 595 1.028 1.623 86.9369 0.548005E+06  
 BETA CL CD CM CROLL CN CY CNTR CMTR CLTR CDTR  
 -0 01 0.91 -1.23 -0.25 0 00 -0.01 0 02 0 47 -0.15 0.46 0.07

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3 375 CP	BP=10.125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP = 12 CP	BP = 16 CP	BP = 22 CP	----TOP----	NO.	CP	
0 0	0.4633	-0.0797	0.0	-0 0974	0.2451	0 1690	-0 3258	-0.4019	38.2	-0.2973	1 -0.4242	
2 5	-0.6771	-1.0300	2 5	-0.4400	0 5877	-0 5923	-0.7825	-1.0109	43.4	-0.2394	2 -0.0940	
5 0	-0 5685	-0 7314	5 0	1.1967	0 2832	-0.5161	-0.6303	-0 8206	50 4	-0.2104	3 -0 4242	
10 0	2 7712	-0 6771	10 0	-0 4780	-0 1736	-0.5541	-0 5161	-0 6683	56 1	-0 1524	4 -0.0940	
15 0	-0 5685	-0 6499	15 0	-0 1355	-0 3258	-0.5161	-0.4400	-0 5541	62.5	-0.1234	5 -0.4242	
25 0	-0 4870	-0 5142	24 0	-0 2877	-0.4019	-0.4780	-0 3639	-0 4400	69 7	-0.1814	6 -0 0940	
35 0	-0 3784	-0 4599	33.0	-0.3141	-0 3141	-0.5342	-0.4242	-0.3141	76.9	-0.1814	7 -0.5265	
50.0	-0 4327		54 0	-0 3141	-0 4242	-0.5342	1 5566	-0 3141	83.4	-0.1524	8 -0 5265	
56 0	-0 4599	-0 4870	65 0	-0 4712	-0 9926	-0 5581	-0.4422	-0 4422	89.4	-0.2104	9 -0 5265	
65 0	-0 5956	-0 6771	78 5	2.1073	-2.3832	-0 8767	-0.5871	-0.5001	95.4	-0.3263	10 -0.5265	
76 0	-0 4599	-1 2744	79.5	131.9177	136.5406	5 0245	26 1734	175.6027	101.1	-0.3263	11 -0 5029	
79 0	-1.8719	-2 0076	80 5	-66.2232	-91.7430	-0 3656	-0.6401	-2 7374	---	BOTTOM---	12 -0.5265	
80.5	-39.2922	-52.4054	81.3	-16.9874	150.0442	-0 4245	-0.4048	-0.4636	38 2	-0.0656		
81.0	5 2989	18.1176	82 0	66.9818	-13 8111	-0.4441	-0.3852	-0 4245	43 4	-0.0366		
82 0	2.9665	-1 2281	84 0	4.9657	0.8104	-0.6989	-0.4832	-0.5029	50.4	-0.1814		
84 0	0.3204		87 0	-9 9307	-4.3838	-0.8753	-0.4636	-0 5225	56.1	-0.0076		
87 0	3 3977		89.0	-22 3634	-2.1300	-0.8801	-0.5029	-0 4557	62.5	-0.0076		
89 0	0 7167	-2 2384	93.0	7.0434	-1.2339	-0 6208	-0.5265	-0.5029	69 7	-0.0656		
93 0	1 1802		96 0	1.4544	-1 5641	-0 4321	-0.2906	-0.3849	76.9	-0.0656		
96 0	2 5998	0 7746	100 0	-3.6628	-2 5309	-0 2199	-0 1727	-0.5029	83.4	-0.0656		
100 0	0 9485	-3 1075	96.0	-0 1963	-0 0312	0 1103	0.1103	0 0631	89 4	-0 0945		
96 0	-0 2683	-0 1814	84 0	0.0631	0.0159	-0 5029	0 1574	1 2422	95 4	-0 0656		
84 0	0 9485	-3.1075	73 5	-0 0940	-0 5342	0 1261	-0 0940	0 1261	101.1	-0.2683		
73 0	-0 2683	-0 1814	54.0	-0 0940	-0 0940	-0 0940	-0 0940	-0.0940				
50 0	0 0214	0 0214	33.0	-0 0940	-0.5342	-0.0940	-0.0940	-0 0940				
35 0	-0 0797	-0 1340	24 0	-0 0974	-0.1355	-0 0974	-0.0593	-0 1736				
25 0	-0 2155	-0 5685	10.0	-0 0974	-0 0974	-0 0593	0.0548	-0 0940				
10 0	-0 1069	-0.1884	5.0	-0.1736	-0.1355	0 0168	0.1310	0 1261				
5 0	-0 1884	-0 2155	2.5	-0 2877	-0 2496	0.0548	0 2832	0.2361				
2 5	0.0561	0 2189										

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TABLE A38 (Concluded)

RUN SEQ 273 6  
 ALPHA 12 06  
 BETA -0 01  
 CLAERO = 0.9809  
 PTOT 2138.28  
 PSTAT 2129 24  
 CL 1.66  
 CD -1 01  
 CDAERO = 0.4274  
 VEL 9 04 87 69  
 CM -0 16  
 CROLL -0.00  
 WING / CANARD PRESSURES  
 DCLC = 0.2187  
 H/C 0.01 2.73  
 CN -0 01  
 CMUC 0.582  
 CY 0 02  
 DWLC = 0.4573  
 CMUW 1.005  
 CNTR 1 06  
 CMUT 1 587  
 CMTR -0 06  
 PAGE 654  
 BASEPR = 8.1904  
 HGT 89 2749  
 RN 0.548478E+06  
 CLTR 1.03  
 CDTR 0.22

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	-5.2392	-2.5508	0 0	0 4736	0 1310	-2.7239	-4 2467	-1.6962	38.2	-0.4132	1	-0 6443
2 5	-2.4422	-2 1164	2 5	-0.5161	-0.5161	-1.6201	-3.7518	-1.5440	43.4	-0.3553	2	0.1261
5 0	-1 7362	-2 1707	5 0	0.0929	-0 6684	-1 2774	-3 5614	-1.5058	50.4	-0.2683	3	-0.7543
10 0	2 6630	-2 0893	10 0	-0.5541	-0 7825	-1 0110	-0 7825	-1.4297	56.1	-0.2104	4	0.1261
15 0	-1 1931	-1 9262	15 0	-0.5161	-0 7825	-0 9348	-0 7825	-1 5820	62.5	-0.2394	5	-0.8644
25 0	-0 8944	-2.0348	24.0	-0 5161	-0 7064	-0 7445	-0.7064	-1.3536	69.7	-0.2394	6	-0.0940
35.0	-0.7585	-1.5733	33 0	-0 4242	-0.5342	-0 6443	-0.5342	-1.5247	76.9	-0.2394	7	-0.5500
50 0	-0 7042		54 0	-0 4242	-0 4242	-0.4242	1.5568	-0.8644	83.4	-0.2104	8	-0.5265
56.0	-0.6771	-0 8129	65.0	-0 6451	-1.1956	-0 6739	-0.5871	-0.8768	89.4	-0.2973	9	-0.5265
65 0	-0 8129	-0.8129	78 5	2 1075	-2.5284	-0 8188	-0.6160	-0 9927	95.4	-0.3553	10	-0.5265
76 0	-0 5142	-1 1931	79 5133	4255 142 4843	5 8876	27 9991	175 8876	101 1	-0 2973	11	-0 5500	
79.0	-1 9805	-1 8176	80.5	-65.6432	-85.2450	-0.4441	-0.6401	-2.4240	---	---BOTTOM---	12	-0.5500
80 5	-39 5321	-52.2542	81 3-20	6935 125 4397	-0 4637	-0.3657	-0.5814	38.2	0.0214			
81 0	6.1032	18.4338	82.0	67.7735	-10.1084	-0.4637	-0 4048	-0.7185	43.4	0.0504		
82 0	3 3001	-0 7185	84.0	4 5154	-0 3852	-0 5225	-0.4637	-0 6009	50 4	-0.1524		
84 0	0 3400		87 0	-6 2464	-3.2276	-0.6205	-0 4245	-0 4832	56.1	0.0793		
87 0	3 3785		89 0-21	.0452 -1 5878	-0 7152	-0.5265	-0 7152	62 5	0.0793			
89 0	0.6878	-2.0357	93 0	5 1574	-1 1397	-0.5029	-0 5029	-0 5500	69 7	0.0793		
93.0	1.0645		96 0	1 4311	-1 3755	-0.3849	-0.3142	-0 6680	76.9	0.0504		
96 0	2.5132	0 8906	100 0	-3.5453	-2.2245	-0.1963	-0.2199	-0 5029	83 4	-0.0076		
100.0	1.1514	-3 0499	96 0	-0 1020	-0 0076	0.1339	0 1103	-0.0076	89.4	0.0214		
96 0	-0 1814	-0.1234	84 0	0.1339	0.0867	-0 5265	0 1575	1 4782	95.4	-0.0076		
84 0	1 1514	-3 0499	73 5	0 0160	-0 4242	0.2361	0 1261	0 3462	101.1	-0.2104		
73 0	-0 1814	-0 1234	54 0	0 0160	0 0160	0 0160	0 0160	0 0160				
50 0	0.1663	0 1373	33 0	0.1261	-0 4242	0 1261	0.1261	0 0160				
35 0	0 0018	-0 0797	24 0	0 0929	0 0929	0 0929	0 2833	0.0929				
25 0	-0.0525	-0 8129	10 0	0 1310	0 1691	0 3213	0 4355	0 3462				
10 0	0 1104	0 0561	5 0	0 2071	0.2071	0 3594	0 5497	0.4563				
5 0	0 0833	0 1104	2 5	0.1691	0.2833	0 4736	0.5117	0 5663				
2 5	0 5449	0.5177										

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RUN SEQ	PROPLULSIVE WING / CANARD PRESSURES										PAGE 655			
274 1	CLAERO = -1 0788	CDAERO = -0 3775	DCLC = 0 2193	DWLC = 1 8754								BASEPR = 8.1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT		RN		
0.01	2138.21	2133 91	4 29	60 38	0 01	2 66	1 262	2.165	3 427	87.0442		0.378549E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR				
-0 01	1.02	-2.70	-0.48	0 00	-0 01	0.02	0.29	-0 28	0 29	0 04				

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**			
BP=3 375 BP=10.125			BP = 2		BP = 6		BP =12		BP =16		BP = 22		----	TOP----
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	0 3153	0 3724	0.0	-1.9637	-0 8419	0.2800	0 4403	0.3601	38.2	-0.3407	1	-0.3889		
2 5	-0 1992	-0 2564	2 5	-0 8419	2.6840	0 4403	-0.2809	-0 5213	43.4	-0 3407	2	-0.1572		
5 0	-0 3136	-0 4279	5 0	3 1648	1 5621	-0 2008	-0 3611	-0.6015	50.4	-0.2797	3	-0.3889		
10 0	3 5735	-0 4279	10 0	-0 8419	-0 0405	-0 2809	-0 2809	-0 5213	56 1	-0.1577	4	-0 1572		
15 0	-0 4851	-0 5422	15 0	-0 2008	-0 3611	-0 5213	-0 3611	-0 6015	62 5	-0.0967	5	-0 3889		
25 0	-0.4279	-0 4851	24 0	-0 5213	-0 5213	-0.6015	-0 3611	-0.5213	69.7	-0.1577	6	-0.1572		
35 0	-0 4279	-0 5994	33 0	-0 3889	-0 3889	-0 3889	-0 3889	-0.3889	76 9	-0.1577	7	-0 7861		
50 0	-0 5422		54 0	-0.3889	-0 6205	-0 6205	0 5379	-0 3889	83.4	-0 2187	8	-0 7364		
56 0	-0 5994	-0 6566	65 0	-0 6456	-1 8655	-0 7676	-0.5236	-0 4627	89 4	-0.1577	9	-0 7364		
65 0	-0 8280	-0 8852	78 5	1 6721	-4 1221	-1 4994	-1 0115	-0 5846	95.4	-0.4627	10	-0 6868		
76 0	-0.7709	-1 9141	79 5268	9001	287 5503	11.2761	56.4204	368.5532	101.1	-0 4627	11	-0 7861		
79 0	-2 7717	-3 1146	80 5*****	-168 3904	-0 3193	-0.8145	-5.0649		---	BOTTOM---	12	-0.7364		
80 5	-72.9447	-112 1469	81 3	7.0672	284 6631	-0 5256	-0 3193	-0 3606	38.2	-0.2187				
81 0	-0 9383	21 1795	82 0128	3452	-24 0867	-0 5256	-0 3606	-0.3606	43.4	-0 0967				
82.0	1.3726	-3 3316	84.0	19 9418	-0 4844	-1.0208	-0 6494	-0 4844	50.4	-0 3407				
84 0	0 9599		87.0	-62 2575	-6 1790	-1 4336	-0.5669	-0 8970	56 1	-0.0967				
87 0	7 7273		89 0	-39 5609	-2 9209	-1 3819	-0.7861	-0 3889	62.5	-0.0967				
89 0	2 0380	-2 9633	93 0	7 1077	-1 8287	-0 9350	-0.7861	-0 7364	69.7	-0.1577				
93 0	2 7090		96 0	-0 9350	-2.3749	-0 6868	-0 3889	-0 2897	76 9	-0.1577				
96 0	5 5757	2 2210	100 0	-7 2402	-4 8572	-0 3393	-0 1407	-0 7364	83.4	-0.2187				
100 0	-0 2187	-5 8300	96 0	-0 2400	-0 0414	0 2068	0 0579	0 1075	89 4	-0 2187				
96 0	-0 6456	-0 5236	84 0	0 2068	0 1075	-0.7364	0 1572	0 0082	95 4	-0.2187				
84 0	-0 2187	-5 8300	73 5	-0 6205	-0 6205	0 3062	0 0745	-0 8522	101.1	-0 5236				
73 0	-0 6456	-0 5236	54 0	-0 1572	-0.1572	-0.1572	-0.1572	-0 1572						
50 0	-0 2187	-0 0967	33 0	-0 1572	-0 6205	-0 3889	-0.1572	-0 1572						
35 0	-0 3136	-0 3136	24 0	-0.2809	-0 4412	-0 3611	-0 2008	-0 3611						
25.0	-0 4851	-0.7709	10.0	-0 2809	-0 4412	-0 3611	-0 0405	-0.1572						
10 0	-0 3707	-0 4279	5 0	-0 4412	-0 7617	-0.3611	-0 0405	-0 1572						
5 0	-0 4279	-0 4279	2 5	-1 0021	-1.2425	-0.5213	-0 2809	0.0745						
2 5	-0.4279	-0 2564												

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TABLE A39 (Continued)

RUN SEQ				PROPLULSIVE		WING / CANARD	PRESSURES			PAGE 656	
274 2	CLAERD = -1 1807	CDAERD = -0 3549	DCLC = 0 1738	DWLC = 1.8188	BASEPR = 8.1904						
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-1 99	2138 21	2133 91	4 29	60 38	0 01	2.67	1.248	2.145	3 393	87.2558	0.378487E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR
-0 01	0.81	-2.73	-0.51	0.00	-0 01	0.03	0.18	-0.31		0.18	0.03

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0 0	-0.1421	0.0866	0.0	-1.9637	-0.8419	0.3601	0.2800	0.2800	38.2	-0.2187	1 -0.3889	
2.5	-0.0849	-0.1421	2.5	-0.7617	2.9244	0.9210	-0.0405	-0.0405	43.4	-0.2797	2 -0.1572	
5 0	-0.1992	-0.1992	5.0	3.2448	1.6422	0.4403	-0.1207	-0.3611	50.4	-0.1577	3 -0.1572	
10 0	3.0019	-0.3136	10.0	-0.8419	0.0396	-0.2809	-0.2008	-0.3611	56.1	-0.0967	4 -0.1572	
15.0	-0.3707	-0.4851	15.0	-0.2008	-0.3611	-0.3611	-0.2008	-0.4412	62.5	-0.0357	5 -0.3889	
25 0	-0.3707	-0.4851	24.0	-0.4412	-0.5213	-0.5213	-0.2809	-0.4412	69.7	-0.0967	6 -0.1572	
35.0	-0.3707	-0.4851	33.0	-0.3889	-0.1572	-0.3889	-0.1572	-0.1572	76.9	-0.0967	7 -0.7364	
50 0	-0.4851		54.0	-0.1572	-0.3889	-0.6205	0.3062	-0.3889	83.4	-0.0967	8 -0.6868	
56.0	-0.5994	-0.5994	65 0	-0.5846	-1.8045	-0.7066	-0.5236	-0.3407	89.4	-0.2797	9 -0.7364	
65.0	-0.8280	-0.8852	78 5	1 0621	-4.0611	-1.4384	-0.9506	-0.4627	95.4	-0.4827	10 -0.7364	
76 0	-0.8280	-1 9141	79.5268	5276	285.3149	12.5965	58.1107	368.7163	101.1	-0.4017	11 -0.7364	
79 0	-2.7717	-3.1717	80.5*****	-170	1250	-0.2368	-0.6907	-4.9410	---	---BOTTOM---	12 -0.7364	
80.5	-72.6140	-112.2703	81.3	9.2542	275.5361	-0.2780	-0.1542	-0.3193	38.2	-0.1577		
81 0	-1.0208	19 7767	82.0	126.5291	-24.5011	-0.4844	-0.2368	-0.1955	43.4	-0.0357		
82 0	1 6201	-3 0015	84.0	20 3131	0.2584	-0.8970	-0.6082	-0.2780	50.4	-0.2797		
84.0	1.0837		87 0	-62.1341	-6 1375	-1.2684	-0.6494	-0.6907	56.1	-0.0967		
87 0	7.7273		89.0	-39 2117	-2.7719	-1.3819	-0.8854	-0.3889	62.5	-0.0967		
89 0	2.0380	-2.9023	93.0	5.8665	-1 8287	-0 9350	-0.7364	-0.6868	69.7	-0.0967		
93.0	2.8309		96 0	-1.0840	-2.4741	-0.6868	-0 4882	-0 2897	76.9	-0.1577		
96 0	5 5757	2 2820	100.0	-7 3892	-4.9067	-0.3393	-0 1904	-0.7861	83.4	-0.1577		
100.0	-0 4017	-5 7080	96.0	-0.2400	-0 0911	0 2068	0 1572	0.0579	89 4	-0.2187		
96 0	-0 5846	-0 5236	84 0	0 1572	0 0579	-0.7364	0.1572	-0.4386	95.4	-0.1577		
84 0	-0 4017	-5 7080	73.5	-0.6205	-0 8522	0 3062	0.0745	-0.3889	101.1	-0.4017		
73 0	-0 5846	-0 5236	54.0	-0 1572	-0.1572	-0 1572	-0.1572	-0.1572				
50 0	-0.1577	-0.0357	33 0	-0.1572	-0.6205	-0 3889	-0 1572	-0.1572				
35 0	-0 3707	-0.4851	24 0	-0.2809	-0.4412	-0.3611	-0.2809	-0.4412				
25 0	-0 4851	-0.8280	10.0	-0 3611	-0.6015	-0 4412	-0.2008	-0 3889				
10 0	-0 4851	-0 4851	5.0	-0 4412	-0.7617	-0 5213	-0 4412	-0 3889				
5 0	-0 4851	-0.5994	2 5	-1.1624	-1 5630	-1.2425	-0.6015	-0.6205				
2 5	-0 8852	-0 8280										

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TABLE A39. (Concluded)

RUN SEQ	PROPLULSIVE WING / CANARD PRESSURES										PAGE 660			
274 6	CLAERO = -0 0779	CDAERO = -0 4764	DCLC = 0.4580	DWLC = 1 9931	BASEPR = 8.1904									
ALPHA	PTOT	PSTAT	Q VEL	YAW H/C	CMUC	CMUW	CMUT	HGT	RN					
12 09	2138 00	2133 59	4.41 61 15	0.01 2.74	1 218	2.095	3.312	89.5629	0.3840	19E+06				
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR				
-0 01	2 37	-2.25	-0.33	-0 01	-0.01	0 03	1.14	-0.13	1.12	0.21				
***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **				
BP=3.375 BP=10 125			BP = 2 BP = 6 BP =12 BP =16 BP = 22					-----TOP----		NO. CP				
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP				
0 0	-5 7396	-2 7328	0 0	-0 0949	0.3734	-2 2023	-4.1536	-1 8122	38 2	-0.4874	1	-1	0331	
2 5	-2 7328	-2 2317	2 5	-0.6412	0 2954	-1 3438	-3 7635	-1 6559	43 4	-0.4280	2	-0	1305	
5 0	-2 0646	-2.3431	5.0	1 6223	-0 0949	-1 2657	-3 6853	-1 5779	50 4	-0 3092	3	-1	0331	
10.0	1 4990	-2 2873	10 0	-0 7974	-0.7974	-1 1876	-0 7974	-1 5779	56.1	-0.2498	4	-0	1305	
15 0	-1 3409	-2 2873	15.0	-0 2510	-0 7193	-1 1096	-0 8755	-1.4999	62 5	-0.2498	5	-1	2588	
25 0	-1.0067	-2 0646	24.0	-0 6412	-0 8755	-0.9535	-0 6412	-1 8122	69 7	-0.2498	6	-0	3561	
35 0	-0 8953	-1 9533	33.0	-0.8074	-0 8074	-1.0331	-0.8074	-1 4844	76.9	-0 2498	7	-0	7431	
50 0	-0 7839		54 0	-0.8074	-0.8074	-0.8074	-1.0331	-1 2588	83.4	-0 1904	8	-0	7914	
56 0	-0 8953	-1.2294	65 0	-0 7845	-1 9133	-0.7845	-0.6657	-0 9033	89 4	-0.2498	9	-0	7914	
65 0	-1.0067	-1.1180	78.5	0.0473	-4 1114	-1 2597	-0 7845	-0 9627	95.4	-0.3685	10	-0	7914	
76 0	-0.6726	-1.6750	79 5261.6572	274 0859	12 2123	57.0693	359.1807	101.1	-0.3685	11	-0	7914		
79.0	-2 9555	-2.6214	80 5*****	-170 4345	-0 4894	-1.0923	-4 6696	---	BOTTOM---	12	-0	7914		
80 5	-71 3939	-108 8543	81.3	9 8408	235.5303	-0 4894	-0.5698	-0 7707	38.2	-0.0121				
81 0	-1.6951	18 5228	82 0122	6277	-26 2951	-0 6903	-0.5295	-0.9315	43 4	0.1067				
82.0	1 0380	-2 5794	84 0	19.6083	1 9224	-0.9315	-0 6501	-0.9315	50 4	-0.1904				
84 0	0 7969		87.0	-59 8567	-7 2823	-1.1726	-0.6501	-0 6903	56.1	0.1067				
87 0	7 3888		89 0	-36 8185	-3 0643	-1 1783	-0.6947	-0 9365	62 5	0.1067				
89 0	1 7702	-2 8044	93 0	4 7217	-1 9521	-0 8881	-0 7431	-0 7914	69 7	0.1067				
93.0	2 6020		96 0	-0.9849	-2 7742	-0 6463	-0.5013	-0 8881	76.9	0.0473				
96.0	5 3349	2 1267	100.0	-6 5461	-4 5152	-0 4045	-0 3562	-0 7914	83 4	-0 0121				
100 0	-0 9033	-5 5968	96 0	-0 3562	-0.1628	0 0307	-0.0177	-0 1628	89 4	-0 0715				
96 0	-0 4280	-0 3685	84 0	0 0307	-0.0177	-0 8398	0 0307	-0 8881	95 4	-0.0715				
84 0	-0 9033	-5 5968	73 5	-0.3561	-0 8074	0 0952	-0 1305	-0 5818	101 1	-0 3092				
73 0	-0 4280	-0 3685	54 0	-0 1305	-0 1305	-0 1305	-0 1305	-0 3561						
50 0	0 1067	0 0473	33 0	-0 1305	-1 0331	-0 1305	-0 1305	-0 1305						
35 0	-0 0601	-0 1714	24 0	0 2173	0 2173	0 0612	0.3734	0 0612						
25 0	-0 0601	-1 0524	10.0	0.2954	0 1393	0 2173	0.4515	0 0952						
10 0	0 0513	-0 0044	5 0	0 2173	0.1393	0 3734	0 6076	0.3209						
5 0	0 0513	0 0513	2 5	0 0612	0 0612	0 3734	0.4515	0 3209						
2 5	0 4967	0.4967												

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TABLE A40. RUN 277, BW6V, DELF=45, CMU=0, (BN/B)W=0.25

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 661				
277 1	CLAERO =	0 5209	CDAERO =	0 2419	DCLC =	0.0	DWLC =	0 0	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.10	2133 40	2103.34	30.06	160.15	0 01	2 67	0.0	0 0	0.0	87.4810	0.100348E+07
BETA	CL	CD	CM	CRDLL	CN	CY	CNTR	CMTR		CLTR	CDTR
-0.01	0.52	0.24	-0.32	0 00	-0 00	0.00	0.52	-0.33		0.52	0.30

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	----	TOP----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	-0.2646	-0 2646	0 0	0.5205	0.4976	0.1198	0.2571	-0 0291	38.2	-0.0880	1	-0.3227
2 5	-0.2565	-0 2646	2.5	-0 2581	-0.3611	-0.5443	-0.5443	-0 7618	43.4	-0.0705	2	0 0745
5 0	-0.2565	-0 2728	5 0	-0 2237	-0.2924	-0.4413	-0.4413	-0.6244	50.4	-0.0705	3	-0 3889
10.0	-0.2728	-0.2728	10 0	-0 2466	-0 3382	-0.4069	-0.4641	-0.5214	56.1	-0.0793	4	0 1076
15 0	-0.2565	-0 2728	15.0	-0 2581	-0 3268	-0.3840	-0 4184	-0 4069	62.5	-0 0967	5	-0.3227
25 0	-0.2646	-0 2728	24.0	-0 2466	-0.3039	-0.3497	-0 3497	-0.3840	69.7	-0 1403	6	0 0083
35 0	-0 2728	-0 2728	33 0	-0 2565	-0 3227	-0 3558	-0 3558	-0 3558	76 9	-0.1490	7	-0.3252
50 0	-0.2646		54.0	-0.2896	-0 3227	-0.4220	0.6703	-0.3558	83.4	-0.1403	8	-0 3110
56.0	-0 2646	-0 2810	65 0	-0 3320	-0.4191	-0.5237	-0.5150	-0 4888	89.4	-0.2187	9	-0.2968
65 0	-0 2646	-0 2810	78 5	0 4523	-0 6544	-0 8026	-0.8287	-0 8461	95 4	-0 2797	10	-0 3110
76 0	-0 2891	-0.2728	79.5	-0 4962	-0 5139	-0.7439	-0 7085	-0 8146	101 1	-0.2361	11	-0.3039
79.0	-0 2810	-0 2646	80 5	-0.4844	-0 5198	-0.7262	-0 6908	-0 2663	----	BOTTOM---	12	-0 3180
80 5	-0 2191	-0 2486	81 3	-0.5080	53.5163	-0 7085	-0 7085	-0.8205	38 2	-0.0619		
81 0	-0 2309	-0 2427	82 0	-0.4962	-1 1212	-0 7320	-0.6908	-0.7969	43.4	-0.0270		
82.0	-0.2309	-0.2545	84 0	-0.5021	-0.5552	-0.6554	-0 7085	-0.8146	50.4	-0.0880		
84.0	-0.2368		87.0	-0.5316	-0.5846	-0.7262	-0.6908	-0.2663	56.1	-0.0008		
87 0	-0.2309		89.0	-0 5805	-0.6373	-0 7791	-0 7578	-0 8430	62.5	0.0253		
89 0	-0 2623	-0 2710	93.0	-0 5663	-0 6443	-0.7437	-0 3252	-0 2968	69.7	0.0602		
93.0	-0 2623		96.0	-0 5592	-0 6656	-0.7437	-0.7437	-0.8713	76.9	0.0689		
96.0	-0.2623	-0 2710	100 0	-0 5167	-0 7011	-0.6514	-0.7295	-0.2897	83 4	0.1473		
100.0	-0 2710	-0 2623	96.0	0.3346	0 3558	0 3346	0.3558	0.2565	89.4	0.1996		
96 0	-0 2710	-0 2623	84.0	0.5332	0.6538	-0.3039	0.6183	0.4977	95 4	0.1212		
84 0	-0 2710	-0 2623	73 5	0 4717	-0 2896	0 6041	0 5710	0.3724	101.1	-0.3145		
73 0	-0.2710	-0 2623	54 0	0 1738	0.2069	0.2400	0 2731	0.1738				
50 0	-0.2797	-0 2710	33 0	-0 0248	-0 2896	0 1076	0.1076	0 0414				
35.0	-0.2728	-0 2810	24 0	0.0511	0 0167	0 0396	0.0739	0.0282				
25 0	-0 2728	-0.2891	10.0	0.0511	0 0396	0.0511	0.0969	0.0414				
10 0	-0 2810	-0 2891	5 0	0 0739	0.0511	0 1083	0.1541	0 1407				
5 0	-0.2646	-0.2810	2 5	0.1083	0 0969	0 1884	0 2228	0 2400				
2 5	-0 2810	-0.2646										

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TABLE A40 (Continued)

RUN SEQ 277 3  
 ALPHA 4 05  
 BETA -0 01

CLAERD = 0 6921  
 PTOT 2133 26  
 PSTAT 2102 86  
 CL 0 69  
 CD 0 29

PROPLUSIVE CDAERD = 0 2946  
 Q VEL 30 40 161 26  
 CM CROLL 0 35 0 00

WING / CANARD DCLC = 0 0  
 YAW H/C 0 01 2 66  
 CN -0 00

PRESSURES DWLC = 0 0  
 CMUC CMUW CMUT 0 0 0 0  
 CY CNTR CMTR 0 00 0.71 -0 36

BASEPR = 8.1904  
 HGT RN 87 0177 0 100600E+07  
 CLTR CDTR 0.69 0.35

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***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP-----	NO	CP	
0 0	-0 2760	-0 2599	0.0	-0 0868	-0 4151	-1.3435	-1.6152	-2 8833	38.2	-0 0857	1	-0.4401
2 5	-0.2599	-0 2518	2.5	-0 2453	-0 8793	-1.0605	-1.2869	-1 6152	43.4	-0.0599	2	0 1818
5 0	-0 2680	-0 2680	5.0	-0 5057	-0.5963	-0 9019	-0.9925	-1.1397	50.4	-0.0771	3	-0.5383
10 0	-0 2599	-0.2680	10 0	-0.2566	-0.5623	-0.7661	-0.7661	-0.8680	56 1	-0.0857	4	0 1818
15 0	-0 2599	-0 2599	15 0	-0.4265	-0 5283	-0.6642	-0 6869	-0.6869	62.5	-0.1202	5	-0.4401
25 0	-0 2680	-0 2599	24 0	-0.3812	-0 4378	-0.5170	-0.5510	-0.5283	69.7	-0.1719	6	0 0181
35 0	-0 2760	-0 2599	33 0	-0.3747	-0.4074	-0.4729	-0.5383	-0 5056	76.9	-0.1719	7	-0.2976
50 0	-0.2599		54 0	-0 3420	-0 4074	-0 4729	0 4763	-0 4729	83.4	-0.1633	8	-0 2976
56 0	-0 2599	-0 2599	65 0	-0 3701	-0.4563	-0 5855	-0.5855	-0 5942	89 4	-0 2150	9	-0.3046
65 0	-0 2680	-0 2518	78 5	0 4658	-0 6631	-0 9044	-0.8871	-0.9388	95 4	-0 2667	10	-0.3046
76.0	-0 2760	-0.2680	79 5	-0.5274	-0.5274	-0 9239	-0 7607	-0 9239	101.1	-0.2236	11	-0 2976
79 0	-0 2599	-0 2680	80 5	-0 5099	-0 5158	-0.8772	-0 7198	-0 2359	---	BOTTOM---	12	-0 3046
80 5	-0 2651	-0 2592	81 3	-0 5099	52.9623	-0 8772	-0 7315	-0.8831	38 2	-0 0254		
81 0	-0 2476	-0 2476	82 0	-0 5216	-1.0521	-0.8656	-0.7432	-0.8772	43.4	0.0004		
82.0	-0 2476	-0 2534	84.0	-0 5216	-0 5508	-0 7607	-0.7315	-0.8831	50.4	-0.0599		
84 0	-0 2359		87 0	-0 5333	-0 5624	-0.8714	-0 7607	-0 2534	56.1	0.0521		
87 0	-0 2534		89 0	-0 6132	-0 6624	-0.9850	-0.8167	-0.9289	62.5	0.0952		
89 0	-0.2322	-0 2581	93.0	-0 6132	-0 6694	-0.8588	-0.3257	-0.2836	69.7	0.1211		
93.0	-0 2581		96 0	-0 5852	-0.6834	-0 8447	-0.8097	-0 9289	76.9	0.1469		
96.0	-0 2495	-0 2409	100 0	-0 5361	-0.7395	-0 7395	-0.8027	-0 2906	83 4	0.1986		
100.0	-0 2495	-0 2581	96.0	0.3687	0 3617	0 3407	0 3477	0 2144	89.4	0.2417		
96.0	-0 2495	-0 2495	84.0	0 5722	0.6563	-0.3046	0.5862	0.4950	95.4	0 1728		
84.0	-0.2495	-0 2581	73 5	0.5091	-0.2765	0 5745	0.5745	0 4109	101.1	-0.3012		
73 0	-0 2495	-0 2495	54.0	0 2145	0 2472	0 3127	0 3127	0 1818				
50 0	-0 2581	-0 2495	33.0	0.0508	-0 3092	0.1818	0.1818	0.0836				
35 0	-0.2680	-0 2680	24 0	0 1622	0 1170	0.1509	0 1962	0.1057				
25 0	-0.2599	-0 2599	10 0	0 1962	0 2075	0.2189	0 3094	0.2145				
10 0	-0 2680	-0 2599	5 0	0 2642	0 2868	0 3547	0.3773	0 3781				
5 0	-0 2680	-0 2599	2.5	0 3773	0 3773	0 4453	0.5019	0 4436				
2 5	-0.2680	-0 2599										

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TABLE A40 (Concluded)

RUN SEQ	277 5	CLAERO = 1.0856	PROPLULSIVE CDAERO = 0.4915	WING / CANARD DCLC = 0 0	PRESSURES DWLC = 0 0	BASEPR = 8.1904
ALPHA	PTOT	PSTAT	0 VEL	YAW H/C	CMUC CMUW CMUT	HGT RN
12.03	2133.12	2103 06	30.06 160.49	0.01 2.75	0.0 0.0 0.0	89.9482 0.998211E+06
BETA	CL	CD	CM	CROLL CN	CY CNTR CMTR	CLTR CDTR
-0 01	1.09	0.49	-0 44	0 01 -0 00	0.00 1.17 -0.45	1 07 0.55

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
%X/C	BP=3 375 CP	BP=10.125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	XLOC	CP	NO	CP
0.0	-0.3088	-0.3006	0.0	-2.8145	-5.5511	-7.0052	-2.9063	-1.5895	38.2	-0.1087	1	-0.6901	
2.5	-0.3006	-0.3006	2.5	-0.2728	-2.2765	-2.6887	-2.7689	-1.5437	43.4	-0.1174	2	0.3360	
5.0	-0.3006	-0.3088	5.0	-1.2461	-1.6239	-2.0933	-2.7346	-1.5323	50.4	-0.1436	3	-1.6832	
10.0	-0.2925	-0.3088	10.0	-0.2843	-1.2575	-1.6124	-2.7117	-1.5208	56.1	-0.1959	4	0.2698	
15.0	-0.2925	-0.3006	15.0	-0.8453	-1.0056	-1.3605	-2.5171	-1.4521	62.5	-0.2482	5	-1.2860	
25.0	-0.3006	-0.2925	24.0	-0.6736	-0.7652	-1.0400	-1.3720	-1.4407	69.7	-0.3266	6	0.0381	
35.0	-0.3088	-0.3006	33.0	-0.6239	-0.6901	-0.8557	-0.9219	-1.4184	76.9	-0.3092	7	-0.3569	
50.0	-0.2925		54.0	-0.4915	-0.5577	-0.6901	0.3029	-1.2860	83.4	-0.2569	8	-0.3498	
56.0	-0.3006	-0.3006	65.0	-0.4835	-0.5619	-0.7536	-0.6752	-1.1545	89.4	-0.3353	9	-0.3569	
65.0	-0.3088	-0.3170	78.5	0.4142	-0.6403	-1.2852	-1.1981	-1.2068	95.4	-0.2743	10	-0.3498	
76.0	-0.3251	-0.3006	79.5	-0.5113	-0.5054	-1.4960	-1.0950	-1.3014	101.1	-0.2133	11	-0.3498	
79.0	-0.3170	-0.3088	80.5	-0.5173	-0.5113	-1.3781	-0.9889	-0.2873	---	---BOTTOM---	12	-0.3427	
80.5	-0.3050	-0.2932	81.3	-0.4877	51.7582	-1.4429	-1.0184	-1.2012	38.2	0.0046			
81.0	-0.2814	-0.2873	82.0	-0.4937	-0.9948	-1.3427	-1.0361	-1.1540	43.4	0.0307			
82.0	-0.2755	-0.2873	84.0	-0.4996	-0.5467	-1.1776	-1.0184	-1.0892	50.4	-0.0390			
84.0	-0.2814		87.0	-0.5173	-0.5821	-1.1835	-1.0007	-0.2873	56.1	0.1092			
87.0	-0.2814		89.0	-0.6051	-0.6406	-1.1442	-1.0591	-1.1159	62.5	0.1876			
89.0	-0.2917	-0.2917	93.0	-0.5696	-0.6548	-0.8889	-0.3781	-0.3498	69.7	0.2486			
93.0	-0.3004		96.0	-0.5696	-0.6619	-0.8038	-0.9740	-1.0875	76.9	0.2486			
96.0	-0.3004	-0.3004	100.0	-0.5484	-0.6974	-0.7328	-0.9101	-0.3356	83.4	0.2747			
100.0	-0.3004	-0.3004	96.0	0.3241	0.4235	0.3809	0.3454	0.1823	89.4	0.2834			
96.0	-0.2917	-0.3004	84.0	0.6646	0.7427	-0.3498	0.6150	0.4093	95.4	0.1701			
84.0	-0.3004	-0.3004	73.5	0.6008	-0.3260	0.6339	0.5677	0.3360	101.1	-0.3266			
73.0	-0.2917	-0.3004	54.0	0.3360	0.3360	0.3691	0.3691	0.2036					
50.0	-0.3004	-0.3004	33.0	0.2367	-0.3260	0.3360	0.3360	0.1705					
35.0	-0.3006	-0.3088	24.0	0.3340	0.3340	0.3569	0.4027	0.2080					
25.0	-0.3088	-0.3006	10.0	0.4828	0.4828	0.4943	0.5286	0.4353					
10.0	-0.3088	-0.3088	5.0	0.5973	0.5630	0.5401	0.5744	0.5015					
5.0	-0.2925	-0.3088	2.5	0.6775	0.5744	0.5057	0.5171	0.4684					
2.5	-0.3006	-0.3088											

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TABLE A41 RUN 280, BW6V, DELF=45, CMU=0 5, (BN/B)W=0.25

RUN SEQ	PROPLUSIVE										WING / CANARD			PRESSURES			PAGE 666	
280 2	CLAERO =	0 7105	CDAERO =	0 4412	DCLC =	0 0	DWLC =	0 4631	BASEPR =	8.1904								
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN							
-0 01	2133.12	2122 94	10 17	93 10	0 01	2 69	0 0	0 535	0 535	87.9400	0 581730E+06							
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR								
-0 01	1 17	0 17	-0.78	0 01	-0 01	0 01	0.81	-0.51	0 81	0 40								
***** CANARD *****			***** WING *****						**FUSELAGE**			**MISC **						
BP=3 375 BP=10.125			BP = 2			BP = 6		BP =12		BP =16		BP = 22		-----TOP----				
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	CP	XLOC	CP	NO	CP				
0 0	-0 3489	-0 3489	0 0	0 5291	0 3938	0.0892	-0.2829	-0 6889	38.2	-0.0788	1	-0 3667						
2 5	-0 3489	-0 3247	2 5	-0 3844	-0 3844	-0.7227	-0.8243	-1.0611	43 4	-0.0788	2	0.1224						
5 0	-0 3730	-0 3247	5 0	-0.3167	-0 3844	-0 6213	-0.7227	-0.8581	50 4	-0 0788	3	-0 5623						
10 0	-0 3489	-0 3489	10.0	-0 4183	-0.3844	-0.6213	-0.5874	-0.7227	56.1	-0.1046	4	0.1224						
15 0	-0 3489	-0 3489	15 0	-0 3506	-0 4521	-0 6213	-0 4859	-0 6889	62.5	-0.0788	5	-0 3667						
25.0	-0 3247	-0 3489	24 0	-0 3844	-0 4183	-0 5536	-0.4521	-0 5874	69.7	-0 1819	6	0.0246						
35 0	-0 3489	-0.3489	33 0	-0.3667	-0 3667	-0 3667	-0 4645	-0 4645	76 9	-0.2076	7	-0 4017						
50 0	-0 3489		54 0	-0 5623	-0 5623	-0 5623	-0.4645	-0 5623	83.4	-0.2076	8	-0.4017						
56 0	-0 3247	-0 3489	65 0	-0 7999	-1 2119	-0 7999	-0 7742	-0 7999	89 4	-0.3106	9	-0 3807						
65.0	-0 3247	-0 3489	78 5	-0 0531	-2 7828	-1 2634	-1 3922	-1 4953	95.4	-0 3106	10	-0 3807						
76 0	-0 3247	-0 3247	79.5-55	6400	-20 7235	-1.2100	-1.3320	-1.5933	101.1	-0 2849	11	-0.3807						
79 0	-0 3247	-0 3489	80.5-48	7406	-1 9940	-1.2623	-1.3320	-0 3214	---BOTTOM---		12	-0 3807						
80 5	-0.3389	-0 3389	81.3-33	0072	156 2230	-1 2448	-1.3145	-1.5585	38 2	-0.0531								
81 0	-0 3389	-0 3389	82 0-22	2572	-3 6492	-1 2971	-1.3320	-1 5759	43 4	-0.0273								
82 0	-0.2866	-0.3737	84 0	1.1421	-1.8546	-1 1229	-1 3668	-1 5759	50.4	-0.1561								
84 0	-0 3389		87 0	-0 1995	-1 6630	-1.3494	-1 3842	-0 3911	56.1	-0 0016								
87.0	-0 3563		89.0	-1 4498	-1 6594	-1.3869	-1 5127	-1 6804	62.5	0 0499								
89 0	-0 3106	-0.3364	93 0	-0.6533	-1.4708	-1 4079	-0 4227	-0 3807	69.7	0.0757								
93 0	-0.3364		96.0	-0.2340	-1 2612	-1.3451	-1 4708	-1.7433	76.9	0.1014								
96 0	-0.3364	-0 3364	100 0	-0 8419	-1.3451	-1 1773	-1 3451	-0.3807	83 4	0 1787								
100.0	-0 3364	-0 3364	96 0	0 5206	0 3739	0 1433	0.2272	0 1014	89.4	0.2302								
96 0	-0 3364	-0 3364	84.0	0 5625	0 6674	-0 3598	0 6464	0.4368	95.4	0.0499								
84 0	-0 3364	-0 3364	73 5	0 5137	-0 3667	0 6115	0 6115	0 4158	101 1	-0.4908								
73 0	-0.3364	-0 3364	54 0	0 2202	0 2202	0 3180	0 2202	0 1224										
50 0	-0 3364	-0 3106	33 0	0 0246	-0.2689	0 1224	0 1224	0.0246										
35 0	-0 3247	-0 3489	24 0	0 0892	0 0554	-0 0123	0 1569	-0 0123										
25.0	-0.3247	-0 3247	10.0	0 0892	0 0554	0.0554	0 1569	0 2202										
10 0	-0.3489	-0 3489	5.0	0.0554	0 0554	0.1569	0 2246	0 2202										
5.0	-0 3489	-0 3730	2 5	0 1569	0 0892	0 2923	0.3261	0 3180										
2 5	-0.3247	-0 3247																

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TABLE A41. (Continued)

RUN SEQ				PROPLULSIVE		WING / CANARD		PRESSURES		PAGE 668	
280 4	CLAERO =	0 8726	CDAERO =	0.5121	DCLC =	0 0	DWLC =	0.4945	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4 08	2133 19	2123.13	10 06	92 61	0 01	2 68	0.0	0 550	0.550	87.6534	0.578059E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	1 37	0.27	-0 83	0 01	-0 01	0.01	1.01	-0.55	0.98	0.47	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.975	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0.0	-0 3628	-0 3628	0.0	-0 1250	-0 6040	-1.8015	-2.3489	-2.9647	38 2	-0.1157	1 -0.4797	
2 5	-0 3383	-0 3383	2 5	-0.3303	-0 8778	-1 2540	-1.5620	-2 4857	43.4	-0.1157	2 0.2127	
5.0	-0.3383	-0 3383	5.0	-0.5698	-0.7066	-1 0145	-1.1172	-1.5278	50.4	-0.1157	3 -0.6775	
10 0	-0 3383	-0.3628	10.0	-0 3303	-0 6040	-0 8778	-0.8778	-1.0830	56.1	-0.1157	4 0.2127	
15 0	-0 3383	-0 3628	15 0	-0 5356	-0 5698	-0.8093	-0 7408	-0.8436	62.5	-0.1678	5 -0.6775	
25 0	-0 3383	-0 3383	24 0	-0 4672	-0.5356	-0 7066	-0 6383	-0.7408	69 7	-0.2460	6 0 0149	
35 0	-0.3628	-0 3628	33.0	-0 4797	-0.4797	-0 5786	-0 6775	-0.6775	76 9	-0.2720	7 -0.4162	
50.0	-0.3628		54 0	-0.5786	-0.6775	-0.6775	-0.3808	-0 6775	83.4	-0.2981	8 -0.3950	
56.0	-0 3628	-0 3871	65 0	-0 8710	-1 3137	-0.8970	-0.9231	-0.9231	89.4	-0.3241	9 -0.3950	
65 0	-0 3628	-0 3628	78.5	-0 0897	-2 8501	-1.3137	-1.4439	-1.6002	95.4	-0.3762	10 -0 4162	
76 0	-0 3628	-0 3383	79.5-55.3584	-20 7556	-1 3040	-1.3392	-1 6388	101.1	-0.3241	11 -0.3950		
79.0	-0 3628	-0 3628	80.5-49.1918	-2.0440	-1.2864	-1.3392	-0.3527	---	---BOTTOM---	12 -0.3738		
80 5	-0 2821	-0.3173	81.3-34.0402	158.0376	-1.2511	-1.3216	-1.5507	38 2	-0.0636			
81 0	-0 3173	-0.3173	82 0-24.0501	-3 4359	-1.2159	-1.3392	-1 6036	43 4	-0.0376			
82 0	-0 3173	-0 3349	84 0 1.0217	-1 8326	-1.1102	-1.3216	-1 6564	50.4	-0.1418			
84 0	-0 3349		87 0 -0 0178	-1 6388	-1.3392	-1.4097	-0 3349	56.1	0.0145			
87 0	-0 2997		89 0 -1.4337	-1.7729	-1.4548	-1.5185	-1.7517	62.5	0.0666			
89 0	-0 3502	-0.3502	93.0 -0 6494	-1.5396	-1.4548	-0.4374	-0.3527	69.7	0.1186			
93 0	-0 3502		96.0 -0.2890	-1 3913	-1.4125	-1.5609	-1.8788	76 9	0.1447			
96.0	-0.3502	-0.3762	100 0 -0 8825	-1.4337	-1.2853	-1.3913	-0 3738	83.4	0.1967			
100.0	-0 3502	-0 3502	96.0 0 5377	0 3469	0.1349	0 1985	0 0077	89.4	0.2488			
96 0	-0.3502	-0.3502	84 0 0.6013	0 7709	-0.3950	0.6649	0 4741	95.4	0.0666			
84.0	-0 3502	-0 3502	73 5 0 6084	-0.2819	0.7073	0 7073	0.4105	101.1	-0.4803			
73.0	-0.3502	-0 3502	54.0 0 3116	0 3116	0.3116	0.3116	0.1138					
50 0	-0 3241	-0.3502	33 0 0.1138	-0.2819	0.2127	0 2127	0.0149					
35 0	-0.3628	-0 3383	24.0 0.2171	0.1829	0.1829	0.2856	0.0803					
25 0	-0.3628	-0 3628	10.0 0.2856	0.2513	0 2513	0.4224	0.3116					
10 0	-0.3628	-0 3383	5.0 0 3882	0 3540	0.4224	0 4908	0 3116					
5 0	-0 3628	-0.3871	2.5 0 4224	0 4566	0 5251	0.5251	0.5095					
2 5	-0.3628	-0 3383										

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TABLE A41. (Concluded)

RUN SEQ	PROPLULSIVE										WING / CANARD		PRESSURES		PAGE 671	
280 7	CLAERO = 1.3745	CDAERO = 0.7977	DCLC = 0 0	DWLC = 0.5143	BASEPR = 8 1904											
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN					
12 07	2133 19	2123.24	9.95	92.09	0 01	2 75	0 0	0.541	0.541	90.0125	0.574659E+06					
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR						
-0 01	1.89	0 63	-0 99	0 01	-0 01	0 00	1.60	-0 71	1.47	0.78						
***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**						
BP=3 375		BP=10 125	BP = 2		BP = 6	BP =12	BP =16	BP = 22	-----TOP----							
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP				
0 0	-0 4263	-0 4016	0.0	-3 6316	-5.5003	-2.8703	-2 2128	-1.7283	38.2	-0.1535	1	-0.7954				
2 5	-0 4016	-0 4263	2.5	-0 4133	-4 9812	-2.9741	-2.0744	-1 8667	43.4	-0.1271	2	0.4052				
5 0	-0 4263	-0 4263	5 0	-1 5553	-1 6591	-3 4585	-2.5588	-1 7976	50 4	-0.2061	3	-1.9959				
10 0	-0 4263	-0 4016	10.0	-0 4480	-1.2784	-2 9395	-2 6973	-1 5899	56.1	-0.2325	4	0.3052				
15 0	-0.4016	-0 4016	15 0	-1 0708	-1 1055	-3.1125	-2.5934	-1.4860	62.5	-0.3379	5	-1.5957				
25 0	-0 4016	-0 4263	24 0	-0 8632	-0 9670	-1.2438	-2.4896	-1 5553	69.7	-0.4169	6	0 0050				
35 0	-0 4016	-0.4263	33.0	-0 7954	-0 7954	-0 7954	-2.1960	-1.5957	76.9	-0 3905	7	-0 4310				
50 0	-0.3770		54.0	-0.7954	-0.8954	-0.7954	-0.3952	-1.7958	83.4	-0.3642	8	-0.4524				
56 0	-0 3770	-0.4016	65 0	-1.2070	-1 7075	-1 0753	-1 2334	-2.0236	89.4	-0.4169	9	-0 4524				
65 0	-0 3770	-0 4016	78 5	-0.1008	-3 9464	-2.6031	-1 7338	-1.7075	95 4	-0 3642	10	-0 4310				
76 0	-0 4016	-0.4263	79 5	-55 4493	-27.7739	-4.4119	-3.5565	-1.7923	101.1	-0.3115	11	-0 4310				
79 0	-0 4016	-0.3770	80.5	-49 6576	-3 8594	-3 6634	-2.6120	-0 4023	---BOTTOM---		12	-0.4310				
80 5	-0.4914	-0 3845	81.3	-36 7561	153 9220	-3.6456	-2.0062	-1.7566	38.2	0.0046						
81 0	-0 4736	-0.4380	82.0	-27 5246	-4 7326	-2.3625	-2.3090	-1.8101	43.4	0.0309						
82 0	-0 4558	-0.3845	84 0	-0.3310	-3 5387	-2 2200	-1.5428	-1.4715	50.4	-0.1008						
84 0	-0 4914		87.0	0.3996	-2.1666	-2.2556	-1.2577	-0.4201	56.1	0.1363						
87 0	-0 4023		89.0	-1 1814	-2.1461	-2.3391	-1.4173	-1.4815	62.5	0.1889						
89 0	-0 4432	-0 4169	93.0	-0.6025	-1 0528	-1 5245	-0 4524	-0.4310	69 7	0.2943						
93 0	-0 4169		96.0	-0.3881	-1 4387	-0 9670	-1.1814	-1 5459	76.9	0.2943						
96 0	-0.4169	-0 4432	100.0	-0.5811	-1 2243	-0 7740	-1 1385	-0 4310	83.4	0.3207						
100 0	-0 4432	-0 4169	96 0	0 5338	0 4695	0.3409	0 2980	0.0192	89.4	0.3207						
96 0	-0.4169	-0 4169	84.0	0.7268	0 8340	-0.4095	0 7268	0.4266	95 4	0.1363						
84 0	-0.4432	-0.4169	73 5	0 8054	-0 3952	0.7053	0.7053	0 5053	101 1	-0.5222						
73.0	-0 4169	-0 4169	54.0	0.4052	0 3052	0 4052	0.4052	0 2051								
50 0	-0 4169	-0.4169	33 0	0 3052	-0.3952	0.4052	0 3052	0 2051								
35 0	-0 4016	-0 3770	24.0	0 4518	0.4518	0 4518	0.5210	0 2788								
25 0	-0.4016	-0 4263	10 0	0 5902	0.5902	0.5902	0 6248	0 4052								
10 0	-0 3522	-0 4016	5 0	0.7286	0 6594	0.6248	0 6594	0.5053								
5.0	-0.4016	-0 4016	2.5	0.7633	0.5556	0.5556	0.5210	0.4052								
2.5	-0.4016	-0 4016														

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TABLE A42. RUN 281, BW6V, DELF=45, CMU=1 0, (BN/B)W=0.25

RUN·SEQ	281 1	CLAERO = 0 6447	CDAERO = 0 5116	DCLC = 0 0	DWLC = 1.0008	BASEPR = 8 1904	PAGE 672
ALPHA	PTOT	PSTAT	0 VEL	YAW H/C	CMUC CMUW	CMUT	HGT RN
-0 07	2133 47	2128.72	4 75 63 57	0 01 2 67	0 0 1.157	1.157	87.4063 0.397499E+06
BETA	CL	CD	CM	CROLL CN	CY	CNTR	CMTR
-0.01	1.65	-0 07	-1 13	0 01 -0 02	0 03	0.86	-0.54
						0.86	0.42

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.3852	-0 4369
2.5	-0 3852	-0 4369
5 0	-0 3852	-0 4369
10 0	-0.3852	-0.3852
15 0	-0.4369	-0 3335
25 0	-0.3852	-0 3852
35 0	-0 4369	-0 3852
50 0	-0 4369	
56.0	-0 4369	-0 4369
65 0	-0 3852	-0.4369
76 0	-0.4369	-0 4369
79.0	-0.3852	-0 4369
80 5	-0.4074	-0.3702
81 0	-0 2955	-0.3702
82 0	-0 4448	-0.4074
84 0	-0 2208	
87 0	-0 2955	
89 0	-0 3684	-0 3684
93 0	-0 4235	
96 0	-0 4235	-0.4235
100.0	-0 4235	-0 4235
96 0	-0 4235	-0 4235
84 0	-0 4235	-0 4235
73.0	-0 4235	-0 4235
50 0	-0 3684	-0 4235
35 0	-0.3852	-0 3852
25.0	-0.4369	-0 4369
10 0	-0 3852	-0.3852
5 0	-0 3852	-0 4886
2 5	-0.4369	-0 3852

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0 0	0 4311	0.3586	-0.5112	-0.5112	-0.8012
2 5	-0 3663	-0 5112	-1 0186	-1 0911	-1 0186
5.0	-0 4388	-0 4388	-0 6562	-0.7287	-1.0186
10 0	-0.3663	-0.5112	-0.7287	-0 6562	-0.8012
15.0	-0.5112	-0 4388	-0.7287	-0 5112	-0.6562
24.0	-0 4388	-0.5112	-0.5837	-0.5112	-0 5837
33.0	-0 6655	-0 6655	-0.6655	-0.6655	-0 6655
54 0	-0 8751	-0.8751	-0 8751	-1.9229	-0 8751
65.0	-0.9752	-1.9684	-0.9201	-0 8649	-0.9752
78.5	-0 9201	-3.5683	-1 5270	-1.5821	-1.7477
79 5*****	-43 7068	-1.4526	-1.5272	-2.0125	101.1 -0.3684
80.5-95.6655	-2 6098	-1 4153	-1.4526	-0.3702	---
81 3-71.2532	333 2561	-1.4526	-1.4899	-1.9006	38.2 -0.0925
82.0-52 7022	-5 6706	-1 3406	-1.3780	-1.7885	43.4 -0.0373
84 0 2 1308	-2 7591	-1 3033	-1.4899	-1 8258	50.4 -0.2029
87 0 2 4667	-2 6098	-1.3780	-1.5272	-0.3328	56.1 0.0178
89.0 -1 2344	-2.5817	-1 5938	-1 6386	-1 9082	62.5 0.0178
93 0 -0 3363	-1 9979	-1 6386	-0 4261	-0 3363	69.7 0.0178
96.0 0 1577	-1.5489	-1 5040	-1 6386	-2 0427	76.9 0.0178
100 0 -1.8633	-1 8633	-1 4141	-1.6837	-0 3812	83 4 0.0730
96 0 0 4721	0 3823	0 0679	0.2475	0.0679	89 4 0.1282
84 0 0.5619	0 7415	-0.3812	0 6517	0.5170	95 4 -0 1477
73 5 0 3823	-0 6655	0.3823	0.5918	0.3823	101.1 -0 5891
54.0 0 1727	0 1727	0 1727	0 1727	-0.0369	
33 0 -0 0369	-0 6655	-0 0369	-0 0369	-0 0369	
24 0 0 1411	0 0686	-0 0039	0 2136	-0 0039	
10 0 0 1411	0 1411	-0 0039	0 2861	-0 0369	
5 0 0.2136	0 1411	0 2136	0 3586	0 1727	
2 5 0 0686	0.1411	0 3586	0.5036	0 1727	

\*\*FUSELAGE\*\*

-----TOP----	XLOC	CP
38.2	-0.1477	
43.4	-0.1477	
50.4	-0.1477	
56.1	-0.2029	
62.5	-0.2029	
69.7	-0.3132	
76.9	-0.3132	
83.4	-0.3132	
89 4	-0.3684	
95 4	-0.3684	
101.1	-0.3684	
---	BOTTOM---	
38.2	-0.0925	
43.4	-0.0373	
50.4	-0.2029	
56.1	0.0178	
62.5	0.0178	
69.7	0.0178	
76.9	0.0178	
83 4	0.0730	
89 4	0.1282	
95 4	-0 1477	
101.1	-0 5891	

\*\*MISC.\*\*

NO.	CP
1	-0.6655
2	-0.0369
3	-0 6655
4	-0 0369
5	-0 6655
6	-0 2465
7	-0.4261
8	-0.3812
9	-0 3812
10	-0.4261
11	-0.3812
12	-0.3812

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TABLE A42. (Continued)

RUN SEQ 281 3  
 ALPHA 4 02  
 BETA -0.01  
 CLAERO = 0 8725  
 PTOT 2133 40  
 CL 1 84  
 CDAERO = 0 5326  
 VEL 4.75 63 55  
 YAW 0.01 2.67  
 H/C 0 0  
 CN 0 03  
 DWLC = 0 9656  
 CMUT 1 074  
 CNTR 1 05  
 CMTR -0 58  
 BASEPR = 8.1904  
 HGT 87 2774 0  
 RN 397797E+06  
 CLTR 1 01  
 CDTR 0.49

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3.375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP-----	NO	CP	
	CP	CP		CP	CP	CP	CP	CP	XLOC			
0 0	-0 3029	-0 3546	0 0	-0 0249	-0 6048	-2 1995	-3 0693	-3.0693	38.2	-0 2791	1 -0 8961	
2 5	-0 3546	-0 3029	2 5	-0 3873	-0 9673	-1.4746	-2.6344	-2.8518	43.4	-0.2791	2 0.1516	
5 0	-0 3029	-0 3029	5 0	-0 6048	-0 7498	-1 2572	-1.6197	-1 7646	50.4	-0.2791	3 -0 8961	
10 0	-0 3029	-0 3546	10 0	-0 3873	-0 7498	-1 0397	-1.1122	-1 2572	56 1	-0.2791	4 -0 0579	
15 0	-0 3029	-0 3029	15 0	-0 6048	-0.6773	-0 9673	-0.8948	-1.2572	62.5	-0.3343	5 -0 8961	
25 0	-0 3546	-0 3029	24 0	-0 5323	-0 6048	-0.7498	-0.8223	-0 8223	69 7	-0.3343	6 -0.2675	
35 0	-0 3029	-0 3029	33 0	-0 6866	-0 6866	-0 8961	-0.8961	-0 8961	76.9	-0 3343	7 -0.4921	
50 0	-0 3546	-0 3546	54 0	-0 8961	-0 8961	-0 8961	-1.7344	-0 8961	83.4	-0.2791	8 -0 5370	
56 0	-0 3546	-0.3546	65 0	-1 0515	-1.9895	-0.9963	-0 9963	-1 1067	89.4	-0 3343	9 -0 5370	
65 0	-0 3546	-0 4580	78 5	-0.9412	-3 5893	-1 5480	-1.6584	-1.8791	95.4	-0.3895	10 -0.5819	
76 0	-0 3546	-0 4580	79 5	*****	-43 3172	-1 5110	-1.5485	-2 2204	101.1	-0.3895	11 -0.5370	
79 0	-0.3546	-0 4063	80 5	-95 1264	-2 6308	-1 5110	-1 5857	-0 6152	---	---BOTTOM---	12 -0 5819	
80 5	-0 3539	-0 2419	81 3	-72.7675	312.5525	-1.4737	-1 5485	-2 0338	38.2	-0.1136		
81 0	-0 4286	-0 4286	82 0	-55 3370	-5 3930	-1 4364	-1 5857	-1.9590	43.4	-0.0584		
82 0	-0 3539	-0.3166	84 0	1 1765	-2 8174	-1 3616	-1 6603	-1.9217	50.4	-0.2239		
84 0	-0 3913		87 0	3 1175	-2 7056	-1 6976	-1.7723	-0 4286	56.1	-0 0584		
87 0	-0 3913		89 0	-1 1658	-2 6476	-1 7496	-1.8844	-2 1089	62.5	-0 0032		
89 0	-0 3895	-0.4446	93 0	-0 4023	-2 1987	-1 8393	-0 4921	-0 4472	69 7	0.1071		
93 0	-0 4446		96 0	0 0917	-1 7945	-1 5251	-1.8393	-2 1987	76.9	0 0519		
96 0	-0 4446	-0 3895	100 0	-1 9741	-1 9293	-1 6148	-1 7048	-0 4472	83.4	0.1071		
100 0	-0.4446	-0 4446	96 0	0 4061	0 3162	-0 0430	0 1366	-0.0430	89.4	0 1071		
96 0	-0 4446	-0 4446	84 0	0.5408	0 7653	-0 4472	0 5857	0 4510	95 4	-0 1688		
84 0	-0 4446	-0 4446	73 5	0 3612	-0 6866	0.3612	0 5708	0.3612	101 1	-0 7205		
73 0	-0 4446	-0 4446	54 0	0 1516	0 1516	-0 0579	0 1516	-0 0579				
50 0	-0 4446	-0 4446	33 0	-0 0579	-0 6866	-0 0579	-0.0579	-0 0579				
35 0	-0 3029	-0 4063	24 0	0 2650	0 1200	0.1925	0 2650	0 1200				
25 0	-0 3546	-0 5097	10 0	0 3375	0 2650	0.3375	0 5550	0.1516				
10 0	-0 3029	-0 4063	5 0	0 4825	0 2650	0 4100	0.5550	0 3612				
5 0	-0 3029	-0 3029	2 5	0 4825	0 4100	0 5550	0.5550	0 3612				
2 5	-0 3029	-0 3029										

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TABLE A42. (Concluded)

RUN SEQ				PROPLULSIVE		WING / CANARD		PRESSURES		PAGE 676
281 5	CLAERO =	1 3798	CDAERO =	0 7991	DCLC =	0 0	DWLC =	0 9847	BASEPR =	8.1904
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT RN
12.20	2133.40	2128 65	4.75	63 55	0 01	2.75	0 0	1.034	1 034	90.0912 0.397730E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0 01	2.36	0.48	-1.31	0.03	-0.02	0 01	1.62	-0.72	1 49	0.79

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0 0	-0.4580	-0.4580	0.0	-3.7941	-5.3164	-3.7216	-2.4168	-1.6921	38 2	-0.1136	1 -0.6866	
2.5	-0.4063	-0.4063	2.5	-0.6048	-5.4613	-3.5767	-2.4168	-1.5472	43.4	-0.1136	2 0.3612	
5.0	-0.4580	-0.5097	5.0	-1.6197	-2.1995	-3.2867	-2.3444	-1.7646	50.4	-0.2239	3 -1.9440	
10 0	-0.4580	-0.4580	10.0	-0.6048	-1.1848	-3.3591	-2.1995	-1.6197	56.1	-0.2239	4 0 5708	
15.0	-0.4580	-0.5097	15.0	-1.1848	-1.1848	-3.4318	-2.8518	-1.5472	62.5	-0.3343	5 -1.5249	
25.0	-0.4580	-0.4580	24.0	-0.9673	-1.1122	-1.6921	-2.7068	-1.9095	69.7	-0.3895	6 -0.0579	
35 0	-0.4063	-0.4580	33.0	-0.8962	-0.8962	-1.1058	-1.9440	-1.5249	76.9	-0.3895	7 -0.4921	
50.0	-0.4580		54 0	-0.8962	-1.1058	-0.8962	-1.3153	-1.7344	83.4	-0.3895	8 -0.4921	
56.0	-0.5097	-0.4580	65.0	-1.3273	-2.3204	-1.3273	-1.0515	-2.0447	89.4	-0.3895	9 -0.4921	
65.0	-0.4580	-0.4580	78.5	-0.7204	-4.8031	-3.5342	-3.1481	-1.5480	95.4	-0.3895	10 -0.4921	
76.0	-0.4580	-0.4580	79.5*****	-49.6251	-3.0788	-3.0788	-1.6230	101.1	-0.2791	11 -0.4921		
79.0	-0.4580	-0.4580	80.5-93	6331	-4.1613	-2.5188	-1.6603	-0.4659	---BOTTOM---		12 -0.4921	
80 5	-0.4285	-0.5032	81	3-73	6647	299.5576	-3.6013	-1.7351	38 2	-0.0032		
81.0	-0.5032	-0.5405	82	0-57	2030	-7.6327	-2.6681	-1.8469	43.4	-0.0032		
82.0	-0.5405	-0.4285	84.0	-0.7272	-4.0120	-2.6681	-1.8469	-1.6603	50.4	-0.1136		
84.0	-0.4659		87.0	3.6400	-3.3027	-2.5936	-1.4737	-0.4285	56.1	0.1071		
87.0	-0.5779		89 0	-0.8514	-3.4560	-1.7048	-1.2555	-1.4800	62.5	0.2175		
89.0	-0.4446	-0.4446	93.0	-0.3574	-2.3783	-1.2555	-0.5370	-0.4921	69.7	0.2726		
93.0	-0.4446		96.0	0.0019	-0.8065	-0.9862	-1.4352	-1.5699	76.9	0.3278		
96.0	-0.4446	-0.4446	100.0	-1.4800	-0.4921	-0.6269	-1.0311	-0.4921	83.4	0.3278		
100 0	-0.4997	-0.4446	96.0	0.5857	0.4959	0.3162	0.2264	0.0468	89.4	0.3278		
96 0	-0.4446	-0.3895	84.0	0.7204	0.8102	-0.4921	0.7204	0.4061	95.4	-0.0032		
81 0	-0.4997	-0.4446	73.5	0.5708	-0.4771	0.5708	0.5708	0.5708	101.1	-0.6653		
73 0	-0.4446	-0.3895	54 0	0.3612	0.5708	0.5708	0.3612	0.1516				
50 0	-0.3895	-0.4446	33.0	0.3612	-0.4771	0.3612	0.3612	0.1516				
35 0	-0.5097	-0.4063	24 0	0.3375	0.4100	0.3375	0.5550	0.1925				
25 0	-0.4580	-0.4580	10.0	0.5550	0.4825	0.4825	0.6274	0.5708				
10 0	-0.4580	-0.4063	5 0	0.6274	0.5550	0.5550	0.5550	0.5708				
5.0	-0.4063	-0.5097	2.5	0.6274	0.4825	0.4825	0.5550	0.3612				
2 5	-0.4063	-0.4063										

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TABLE A43 RUN 282, BW6V, DELF=45, CMU=2 0, (BN/B)W=0.25

RUN SEQ	PROPLULSIVE										WING / CANARD		PRESSURES		PAGE 679	
282 1	CLAERO =	0 6069	CDAERO =	0 5751	DCLC =	0.0	DWLC =	1 7321	BASEPR =	8.1904						
ALPHA	PTDT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN					
0 02	2133 47	2130 76	2 71	48.00	0 01	2 66	0 0	2 000	2 000	87.0273	0.301292E+06					
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR						
-0 01	2 34	-0 42	-1.63	0 03	-0 03	0.05	0 95	-0.60	0 95	0 43						

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP	
	CP	CP		CP	CP	CP	CP	CP	XL0C			
0.0	-0 3621	-0 4526	0.0	0 3607	-0 0197	-0.6537	-0.6537	-1.2878	38.2	-0 0669	1 -0.4717	
2.5	-0 4526	-0.4526	2 5	-0.4001	-0 5270	-1.2878	-1.0342	-1.7950	43 4	0.0296	2 0.2616	
5.0	-0 4526	-0.4526	5 0	-0 4001	-0.4001	-0 9074	-0 6537	-1.1610	50.4	-0.0669	3 -0.4717	
10.0	-0 3621	-0 4526	10.0	-0 4001	-0 5270	-0.7806	-0 5270	-0.9074	56.1	-0.0669	4 0 2616	
15 0	-0 4526	-0.5430	15 0	-0 4001	-0 5270	-0 9074	-0.5270	-0.7806	62.5	-0.0669	5 -0.4717	
25.0	-0.4526	-0 4526	24 0	-0.5270	-0.6537	-0 6537	-0.4001	-0 9074	69.7	-0.0669	6 -0.1051	
35.0	-0.4526	-0 3621	33.0	-0.1051	-0 4717	-0 4717	-0.4717	-0.4717	76.9	-0.1635	7 -0.4194	
50 0	-0 4526		54 0	-0 4717	-0.4717	-0.8383	-2 6715	-0.8383	83.4	-0.1635	8 -0 4194	
56 0	-0 4526	-0 4526	65 0	-0 9357	-2 4800	-0.9357	-0 9357	-1.1287	89 4	-0.2600	9 -0 4194	
65 0	-0 4526	-0 4526	78.5	-1.9975	-4.2176	-1.3218	-1 6114	-3.8315	95.4	-0.1635	10 -0.4194	
76 0	-0.4526	-0.4526	79.5*****	-69 9916	-1.3561	-1 8132	-3.6416	101.1	-0.2600	11 -0 3409		
79 0	-0 3621	-0 4526	80 5*****	-4 1642	-1 5521	-1 5521	-0.1806	---BOTTOM---	12 -0 4194			
80.5	-0 3766	-0 3113	81 3*****	568 0386	-1.2908	-1 8785	-2.9235	38 2	0.0296			
81 0	-0.1153	-0 3766	82.0-89.9739	-8 4089	-1 2908	-2.4010	-3 5112	43 4	0.0296			
82 0	-0 3113	-0 4419	84.0 7 9822	-3 5764	-1.4867	-1.9438	-2 7927	50.4	-0.0669			
84 0	-0 3766		87.0 5 1088	-3 6416	-1.4867	-1 9438	-0.2459	56.1	0 1260			
87 0	-0.2459		89.0 -1.9908	-3 3262	-1 9122	-2.5408	-2 3836	62.5	0 1260			
89 0	-0.2600	-0.3565	93.0 -0.1837	-2 3836	-2 3836	-0.4980	-0 4980	69 7	0.1260			
93.0	-0.2600		96.0 0 4448	-1 6765	-1 9908	-1.9908	-1 7550	76 9	0.1260			
96 0	-0 2600	-0.2600	100 0 -3 3262	-2 0693	-1.2050	-0 8122	-0 3409	83.4	0.2226			
100 0	-0 3565	-0 2600	96 0 0 4448	0 2878	0 0521	0 2878	0.1306	89 4	0.1260			
96 0	-0.2600	-0 2600	84 0 0 6020	0.7592	-0.4194	0 7592	0 4448	95.4	-0 1635			
84 0	-0.3565	-0 2600	73 5 0.6282	-0 4717	0 6282	0.6282	0 6282	101.1	-0.6461			
73 0	-0 2600	-0 2600	54 0 0 2616	0 2616	0 2616	0.2616	0 2616					
50.0	-0 3565	-0.2600	33 0 0 2616	-0.4717	0 2616	0 2616	0 2616					
35 0	-0 4526	-0.4526	24 0 0 2340	-0 1465	-0.0197	0.3607	-0 0197					
25 0	-0 4526	-0 4526	10 0 0 1071	0.1071	0 1071	0.3607	0.2616					
10 0	-0 3621	-0 4526	5.0 0 1071	0 1071	0 1071	0 4876	0 2616					
5 0	-0 5430	-0.4526	2 5 0 1071	0 1071	0 3607	0 3607	0 6282					
2 5	-0 4526	-0 4526										

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TABLE A43. (Continued)

RUN SEQ				PROPLUSIVE		WING / CANARD		PRESSURES		PAGE 681
282 3	CLAERO =	0 6805	CDAERD =	0 6649	DCLC =	0 0	DWLC =	1 9312	BASEPR =	8.1904
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	RN
4.02	2133 54	2130 94	2.60	46 98	0.01	2 68	0.0	2.148	2.148	87.7956 0 295076E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0 01	2.61	-0.28	-1.72	0 02	-0 03	0.04	1.13	-0.64	1.10	0.51

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
%X/C	BP=3 375	BP=10 125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP-----	NO.	CP	
	CP	CP		CP	CP	CP	CP	CP	XLOC			
0 0	-0.4164	-0.3219	0 0	-0.4560	-1.3823	-3 2351	-2.8380	-2 5734	38.2	-0.3097	1	-0.5307
2 5	-0.4164	-0.4164	2.5	-0.4560	-1.3823	-2 0440	-2.0440	-2.9705	43.4	-0.2090	2	0.2346
5 0	-0.4164	-0.4164	5.0	-0.8530	-0.9853	-1.3823	-1.5146	-1 6470	50.4	-0.2090	3	-0.5307
10 0	-0.3219	-0.4164	10 0	-0.4560	-0.8530	-1 1176	-1.1176	-1.5146	56.1	-0.3097	4	0.2346
15.0	-0.3219	-0.4164	15.0	-0.7206	-0.7206	-0.9853	-0.7206	-0.9853	62.5	-0.4104	5	-0.9133
25 0	-0.4164	-0.3219	24 0	-0.5883	-0.7206	-0.7206	-0.5883	-0.8530	69.7	-0.4104	6	-0.1480
35 0	-0.4164	-0.3219	33 0	-0.5307	-0.5307	-0.5307	-0.9133	-0.5307	76.9	-0.5112	7	-0.5581
50 0	-0.4164		54 0	-0.9133	-0.9133	-0.9133	-2 4438	-0.9133	83.4	-0.5112	8	-0.3941
56 0	-0.4164	-0.4164	65 0	-1 3170	-2.9285	-1 2163	-1.3170	-1 3170	89.4	-0.5112	9	-0.4761
65 0	-0.3219	-0.3219	78.5	-2.2235	-4.8424	-1 8206	-1 8206	-3.7345	95.4	-0.5112	10	-0.4761
76 0	-0.3219	-0.4164	79.5*****	-76 6900	-1.7261	-2.0669	-2 9528	101.1	-0.4104	11	-0.4761	
79 0	-0.3219	-0.3219	80.5*****	-3.4299	-1 7261	-1.7261	-0.6358	---BOTTOM---	12	-0.4761		
80.5	-0.5677	-0.6358	81 3*****	555 0640	-1.7261	-1.8625	-3.0893	38.2	-0.1082			
81 0	-0.5677	-0.4995	82 0-95	1588	-9.0184	-1.7943	-2.3395	43.4	-0.2090			
82 0	-0.4995	-0.4314	84 0 7	3375	-4 1113	-1.5218	-1.6581	50.4	-0.3097			
84 0	-0.4995		87.0	5 3614	-3.7706	-1.8625	-1 7943	56.1	-0.1082			
87 0	-0.3632		89.0	-2 0339	-3.9199	-2.1979	-2.4438	62.5	-0.1082			
89 0	-0.5112	-0.5112	93 0	-0.3121	-2.9359	-2 1979	-0.5581	69.7	0 0932			
93.0	-0.5112		96.0	0 1799	-2.0339	-1 8699	-2.1979	76.9	-0.0075			
96 0	-0.5112	-0.5112	100 0	-3 5096	-2.4438	-1 7060	-1.9519	83.4	-0.0075			
100 0	-0.5112	-0.5112	96 0	0.3439	0 2619	-0.1480	0.1799	89.4	-0.0075			
96 0	-0.5112	-0.5112	84 0	0 5079	0.7538	-0.4761	0.6718	95.4	-0.3097			
84 0	-0.5112	-0.5112	73 5	0.6172	-0 1480	0.2346	0.6172	101.1	-0.9141			
73.0	-0.5112	-0.5112	54 0	0 2346	0 2346	0.2346	0 2346	0 2346				
50 0	-0.5112	-0.6119	33.0	0 2346	-0.5307	0.2346	0.2346	0.2346				
35.0	-0.4164	-0.3219	24 0	0 0734	0 2058	-0 0590	0 4704	0 0734				
25 0	-0.4164	-0.3219	10 0	0 3381	0.0734	0 2058	0.4704	0 2346				
10.0	-0.4164	-0.3219	5.0	0 4704	0.4704	0 4704	0 6027	0 2346				
5.0	-0.3219	-0.4164	2 5	0 6027	0 6027	0.4704	0.6027	0 6172				
2 5	-0.3219	-0.4164										

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TABLE A43. (Concluded)

RUN SEQ				PROPLULSIVE		WING / CANARD		PRESSURES		PAGE 683	
282 5	CLAERO =	1 2496	CDAERO =	0 8061	DCLC =	0.0	DWLC =	1 8461	BASEPR =	8 1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12 13	2133 54	2130 94	2.60	46 98	0 01	2 74	0.0	1.940	1.940	89.7773	0.295104E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0 01	3 10	0 21	-1.81	0 03	-0.03	0.02	1.61	-0.73	1.48	0.78	

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
%X/C	BP=3 375	BP=10.125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO.	CP	
	CP	CP		CP	CP	CP	CP	CP				
0 0	-0 5107	-0.5107	0 0	-3 7645	-5.0877	-3.2351	-2 3087	-1.9116	38 2	-0.1082	1 -0.5307	
2 5	-0 5107	-0 4163	2 5	-0 5883	-4.9554	-3.3673	-2.1764	-1.7793	43.4	-0.2090	2 0 6172	
5 0	-0 5107	-0.6050	5 0	-1 7793	-4 6908	-2.7057	-2.5734	-2 0440	50 4	-0.3097	3 -1.6785	
10.0	-0 5107	-0 6050	10 0	-0 5883	-1.1176	-3 4996	-2.4410	-2.0440	56.1	-0.3097	4 0.6172	
15.0	-0 6050	-0 6050	15 0	-1 1176	-1.1176	-3.6323	-2 1764	-2 0440	62.5	-0.3097	5 -1.6785	
25 0	-0.5107	-0 5107	24 0	-0 9853	-0 9853	-1.9116	-2.3087	-2.3087	69.7	-0.5112	6 0 2346	
35 0	-0 5107	-0 6050	33 0	-0 5307	-0 9133	-1 6785	-2.0610	-1.6785	76.9	-0.4104	7 -0.4761	
50.0	-0.5107		54 0	-0 9133	-0.9133	-0 9133	-2 8263	-2 0610	83.4	-0.4104	8 -0.5581	
56 0	-0 6050	-0 6050	65 0	-1.4178	-3 2308	-1 4178	-1.3169	-1 5185	89.4	-0.5112	9 -0.5581	
65 0	-0 5107	-0.5107	78 5	-2.1228	-5 8499	-3.2308	-1.7199	-1 3169	95.4	-0.4104	10 -0.5581	
76 0	-0.4163	-0 6050	79 5	*****	-95 0204	-5 0654	-6.1559	-1.5218	101.1	-0.4104	11 -0.5581	
79 0	-0 5107	-0.6050	80 5	*****	-6 4285	-3 8387	-4.1113	-0 4995	---	BOTTOM---	12 -0.5581	
80 5	-0.5677	-0 4314	81 3	*****	535 8376	-1.9988	-1.6580	-1.2491	38.2	0.0932		
81 0	-0 4995	-0.3632	82 0	*****	-11.6077	-2 4759	-1.5218	-1.5218	43.4	0.0932		
82 0	-0 3632	-0 4995	84 0	6 9969	-5 4061	-2.6122	-1 3855	-1.4536	50.4	-0.2090		
84 0	-0 4995		87 0	6 2474	-4.5886	-3.1573	-1.2491	-0 6358	56.1	0.0932		
87 0	-0 5677		89 0	-1 7879	-2.7719	-1.6239	-1.5420	-1.5420	62.5	0.2947		
89 0	-0 4104	-0 5112	93 0	-0 4761	-1.7879	-1 4600	-0 5581	-0 5581	69.7	0.1940		
93 0	-0.4104		96 0	-0.0661	-1.2960	-1.2140	-1.4600	-2 2799	76 9	0.2947		
96 0	-0 4104	-0 4104	100 0	-2.9359	-1 2960	-1 0500	-1.4600	-0 6401	83 4	0.1940		
100 0	-0.5112	-0 3097	96 0	0 4258	0 2618	0 0979	0 1799	-0 0661	89.4	0.1940		
96 0	-0 5112	-0 3097	84 0	0.5079	0.7538	-0 5581	0 5898	0 4258	95 4	-0.1082		
84 0	-0 5112	-0 3097	73 5	0 9998	-0 1480	0 6172	0 9998	0 2346	101.1	-0.9141		
73 0	-0 5112	-0 3097	54 0	0 6172	0 2346	0 6172	0.6172	0.2346				
50 0	-0 5112	-0 4104	33 0	0 2346	-0 1480	0.2346	0 2346	0 2346				
35 0	-0 5107	-0.5107	24 0	0 2058	0 2058	0 2058	0 4704	-0 0590				
25 0	-0 5107	-0 5107	10 0	0 4704	0 4704	0.4704	0 6027	0.2346				
10 0	-0.5107	-0 5107	5 0	0 6027	0 6027	0 6027	0.6027	0.6172				
5 0	-0.5107	-0 6050	2 5	0 7351	0 6027	0 6027	0.4704	0 2346				
2.5	-0 5107	-0.5107										

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RUN SEQ	289 2	CLAERO =	0 5316	CDAERO =	0 2390	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1904	PAGE 711
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
-0 06	2133 19	2103 01	30 17	160 35	-0 05	2 66	0 0	0.0	0.0	87.0440 0	100699E+07	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR	
0 05	0.53	0 24	-0 18	0 00	-0 00	0 00	0 52	-0 17		0.52	0 27	

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10.125
%X/C	CP	CP
0 0	0.5127	0 5452
2 5	-0 0895	-0 2522
5.0	-0.1383	-0 2603
10 0	0 7242	-0 2847
15.0	-0.2441	-0 2929
25 0	-0.2441	-0 2685
35 0	-0.2359	-0.2441.
50 0	-0.2359	
56 0	-0.2522	-0.2685
65 0	-0 3173	-0 3417
76 0	-0 2603	-0.5207
79 0	-0.4882	-0.6590
80 5	-0 4852	-0.6496
81 0	-0.4617	-0 6261
82.0	-0.4734	-0.6496
84 0	-0.2796	
87.0	-0.4910	
89.0	-0.4837	-0 6834
93 0	-0 4924	
96 0	-0 5010	0.6189
100.0	0.7491	-0 6486
96 0	0 3411	0 3411
84 0	0.7491	-0 6486
73 0	0 3411	0 3411
50.0	0 6536	0 6102
35 0	0 5289	0 5452
25 0	0 1709	-0 3336
10 0	0.0407	0 0407
5.0	-0 0895	-0 0081
2 5	-0.1790	-0 0488

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	0 0542	0.1112	0.2937	0.0998	0 0770
2.5	-0.2310	0.2025	-0.2082	-0.4363	-0 6872
5.0	0.1454	0 0998	-0.2310	-0 3793	-0 5732
10.0	-0 2424	-0 0371	-0.2766	-0 3907	-0 5047
15.0	-0 0143	-0.1055	-0.2994	-0.3222	-0.4249
24.0	-0 0941	-0.1739	-0.2994	-0.2994	-0.3564
33.0	-0.1533	-0.2192	-0.3511	-0.3511	-0.3511
54 0	-0.2522	-0 3182	-0.4171	0 1765	-0 3511
65 0	-0 3101	-0.3622	-0 5097	0 4576	-0 4490
78.5	0.1587	-0.6313	-0.8830	-0.7702	-0.7789
79.5	-0.5028	-0.5263	-0.8904	-0 6908	-0.7671
80.5	-0.5439	-0.5380	-0.8963	-0.7025	-0.2796
81.3	-0.4969	53.2363	-0.7965	-0.6614	-0 7495
82.0	-0 5263	-1.1606	-0.8435	-0.6438	-0.7436
84.0	-0.5204	-0.5615	-0.6908	-0.6908	-0 7612
87.0	-0.5615	-0.5909	-0 7965	-0 6849	-0.2913
89 0	-0.5891	-0 6245	-0 8718	-0 7163	-0 7588
93.0	-0 5750	-0 6457	-0 7870	-0.2994	-0.2711
96.0	-0.5609	-0 6527	-0 7588	-0 7022	-0.7941
100.0	-0.5185	-0 6598	-0.6810	-0.6952	-0.2641
96 0	0.2165	0.2871	0.3295	0 3790	0.2801
84.0	0.4073	0 5274	-0.2782	0 5840	0 5204
73.5	0 3414	-0 2522	0.4733	0 5062	0 4733
54.0	0 1435	0 1435	0 2095	0.2424	0.1765
33 0	0 0116	-0 2522	0.0775	0.0775	0 0446
24 0	-0 0371	-0 0143	0 0428	0 0884	0 0199
10 0	-0.1055	-0.1055	0 0085	0.0998	0 0775
5 0	-0 1169	-0.1625	0.0085	0.1568	0.1435
2 5	-0 1511	-0.2880	-0 0257	0 1682	0.2095

\*\*FUSELAGE\*\*

	----	TOP----	**MISC.**
XLOC	CP	NO.	CP
38.2	-0.1625	1	-0.2522
43.4	-0.1538	2	0.0446
50.4	-0.1191	3	-0.3841
58.1	-0.0843	4	0.0775
62.5	-0 0583	5	-0.2852
69.7	-0.0843	6	0.0116
76.9	-0 0930	7	-0.2923
83.4	-0.0930	8	-0.2782
89.4	-0.1972	9	-0.2782
95.4	-0.2408	10	-0.2853
101.1	-0.2059	11	-0.2853
	---	BOTTOM---	
38.2	0.0025		
43.4	0.0285		
50.4	-0.0583		
56.1	0.0198		
62.5	0.0112		
69.7	0.0025		
76.9	0 0459		
83 4	0.1067		
89.4	0.1587		
95.4	0.0719		
101.1	-0.3101		

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TABLE A44 (Continued)

RUN SEQ 289 4  
 ALPHA 4 15  
 BETA 0 05

CLAERO = 0 7639  
 PTOT 2133 33  
 CL 0 76

CDAERO = 0 3084  
 PSTAT 2103.04  
 CD 0 31

PROPLULSIVE  
 Q VEL 30 29 160 84  
 CM -0 16

WING / CANARD  
 DCLC = 0 0  
 YAW H/C 2 67  
 CN -0 00

PRESSURES  
 DWLC = 0.0  
 CMUC CMUW CMUT 0 0 0 0  
 CY CNTR CMTR 0 00 0.77 -0 14

PAGE 713  
 BASEPR = 8.1904  
 HGT RN 87 2514 0 100584E+07  
 CLTR CDTR 0.75 0.33

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC ***	
%X/C	BP=3 375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	-----TOP----	NO	CP
0 0	0 0844	-0 6127	0 0	0 1255	0 2391	-0 3859	-1 1700	-2 6587	38.2 -0 2018	1	-0 3794
2 5	-0 7425	-1 0181	2 5	-0 2609	-0 0336	-0 7041	-1 0109	-1.5905	43 4 -0.1932	2	0 1463
5 0	-0 5884	-0 6857	5 0	-0 0336	-0 1132	-0 5336	-0 8632	-1 1814	50.4 -0.1499	3	-0 4451
10 0	0 6195	-0 6209	10 0	-0 2609	-0 2268	-0 4996	-0.6473	-0 8064	56.1 -0 1067	4	0 1791
15 0	-0 4912	-0 5722	15 0	-0 1586	-0 2609	-0 4427	-0 5564	-0 6586	62 5 -0.0894	5	-0.4123
25 0	-0 3858	-0 4344	24 0	-0.2268	-0.3177	-0 4314	-0 4654	-0.5223	69 7 -0 1326	6	0 0149
35 0	-0 3209	-0.3533	33 0	-0 2480	-0 3137	-0.4123	-0.4451	-0.4123	76 9 -0.1413	7	-0.3161
50 0	-0 3209		54 0	-0 3137	-0.3465	-0 4780	0.1134	-0.4123	83.4 -0.1326	8	-0.3020
56 0	-0 3209	-0 3452	65 0	-0.3662	-0.4526	-0 6170	-0 5651	-0 5564	89 4 -0.2105	9	-0 3020
65 0	-0 3776	-0.4425	78 5	0 1874	-0 6429	-1 0841	-0 9543	-0.8332	95.4 -0.2451	10	-0 3020
76 0	-0.2804	-0 6127	79.5	-0.5678	-0 5444	-1.2116	-0.8780	-0.8253	101.1 -0.1932	11	-0 2950
79 0	-0.5398	-0 7263	80 5	-0.5444	-0.5678	-1 1940	-0 8604	-0 2518	---BOTTOM---	12	-0 2950
80 5	-0 5386	-0 7200	81 3	-0 5327	53 1121	-1 1062	-0 8604	-0.7961	38.2 0 0404		
81 0	-0.5152	-0 7024	82 0	-0 5386	-1 1121	-1 1413	-0 8838	-0 8312	43 4 0 0577		
82 0	-0 5620	-0 7200	84 0	-0 5503	-0 5912	-1 0360	-0 9014	-0.7961	50 4 -0.0288		
84 0	-0 3045		87.0	-0 6029	-0.6088	-1.1706	-0 8780	-0 2694	56.1 0.0490		
87 0	-0 5327		89.0	-0 6470	-0 6682	-1 1610	-0 9286	-0 8230	62 5 0 0663		
89 0	-0 5478	-0 7208	93 0	-0 6330	-0 6682	-0 9639	-0 3161	-0 2950	69 7 0 0750		
93 0	-0 5391		96 0	-0 6259	-0 6611	-0 8653	-0.9146	-0 8864	76.9 0.1009		
96 0	-0 5391	0 7064	100 0	-0.5837	-0 6470	-0.7245	-0 9146	-0.2880	83.4 0.1615		
100 0	0 6891	-0 7035	96 0	0 2612	0 3316	0 3598	0.3527	0.2471	89.4 0 1961		
96 0	0 3431	0 3345	84 0	0 4443	0 5851	-0 3020	0.5851	0 4583	95.4 0 0923		
84 0	0 6891	-0 7035	73 5	0.4420	-0 2480	0 5734	0.5405	0.4420	101.1 -0.3143		
73 0	0 3431	0 3345	54 0	0 2448	0.2448	0 3106	0.3106	0.2120			
50 0	0 6286	0 6632	33 0	0 1134	-0 2480	0 1791	0 1791	0.1134			
35 0	0 5708	0 5627	24.0	0 0459	0 0800	0 1141	0.1709	0.1028			
25 0	0 2141	-0 4101	10 0	0.0118	0.0459	0 1596	0 2618	0.2448			
10 0	0 1249	0 1249	5 0	0 0005	-0 0109	0 2164	0.3187	0 3763			
5 0	0 0115	0 1249	2 5	-0 0677	-0 0336	0 2732	0 3982	0 4420			
2 5	0 2385	0 3763									

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TABLE A44. (Concluded)

RUN SEQ	PROPLULSIVE	WING / CANARD	PRESSURES	PAGE 715
289 6	CLAERO = 1.2544 CDAERO = 0.5250 DCLC = 0 0 DWLC = 0.0		BASEPR = 8.1904	
ALPHA	PTOT PSTAT Q VEL YAW H/C CMUC CMUW CMUT		HGT RN	
12 15	2133.47 2103.64 29.83 159.73 -0.04 2 72 0.0 0.0 0 0		89.0004 0 996620E+06	
BETA	CL CD CM CROLL CN CY CNTR CMTR		CLTR CDTR	
0.04	1.25 0 52 -0 10 -0 00 -0.00 0.00 1.32 -0.08		1.23 0.55	

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3 375	BP=10.125
%X/C	CP	CP
0.0	-5.8034	-1.6394
2.5	-2.4459	-1.5407
5.0	-1.7711	-1.4748
10.0	0 5331	-1 4584
15.0	-1.1292	-1 4419
25 0	-0.8082	-1.4666
35 0	-0 6683	-1.3596
50.0	-0 5285	
56.0	-0.5120	-1.0222
65 0	-0.5531	-0.8247
76.0	-0.3392	-0.8412
79.0	-0 6766	-0 8905
80.5	-0 6908	-0.9463
81 0	-0.6611	-0.9284
82.0	-0 6611	-0 9284
84.0	-0.3819	
87.0	-0.6849	
89 0	-0.6748	-0.9207
93.0	-0.6661	
96 0	-0.6748	0.7037
100.0	0.6422	-0.8504
96.0	0 3349	0.3086
84.0	0.6422	-0.8504
73.0	0.3349	0.3086
50 0	0 8090	0.6422
35.0	0.6813	0.5825
25 0	0.3356	-0.5778
10 0	0.3109	0.2698
5 0	0.2780	0.3109
2.5	0.6236	0 5825

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0 0	0.0217	-1 0627	-4.9849	-3.7390	-1 5933
2 5	-0.3244	-0 7974	-1.9971	-3.7390	-1.6279
5.0	-0 4398	-0 7628	-1.4318	-3.8889	-1 6164
10 0	-0.3129	-0.6705	-1.1088	-1.4895	-1.5703
15 0	-0 5090	-0.6474	-0.9589	-0.7858	-1 5587
24.0	-0.4974	-0 5782	-0 7397	-0.7166	-1.5703
33 0	-0.4986	-0 5653	-0 6654	-0.6654	-1.4991
54 0	-0 4652	-0 5319	-0.6320	0 0350	-1.1656
65.0	-0 4992	-0.5783	-0.7363	-0.7363	-0 9470
78.5	0.1856	-0 6661	-1 4475	-1.3070	-1.2631
79.5	-0 5839	-0 5958	-1.8433	-1.1601	-1.3562
80.5	-0.6017	-0.5780	-1 7542	-1.1364	-0.3106
81 3	-0 6136	53 9150	-1 6473	-1 1483	-1.1898
82.0	-0.5601	-1 1007	-1 7007	-1.1780	-1 1898
84 0	-0 6017	-0 6314	-1.3977	-1.0888	-1.1245
87.0	-0.6077	-0.6671	-1.2077	-1.1601	-0.3225
89 0	-0.6774	-0.7203	-1.1133	-1.1920	-1.1133
93.0	-0.7131	-0.7131	-0.8918	-0.3915	-0.3486
96.0	-0.6559	-0 6845	-0.8560	-1.1062	-1.1062
100.0	-0.6059	-0 6130	-0.7417	-1 0204	-0.3486
96 0	0.3232	0 4090	0 3875	0.3589	0.2017
84 0	0.6020	0 7092	-0 3557	0.5877	0.3804
73.5	0.5686	-0.3318	0 6353	0.5686	0.4019
54 0	0.3352	0.3352	0 3685	0 3685	0.2351
33 0	0.2351	-0 3318	0.3018	0 3352	0.2018
24.0	0 1947	0 2524	0 3101	0.3677	0.2178
10.0	0 2178	0.2870	0 4254	0 4716	0.4352
5 0	0 2524	0.3447	0 4600	0.5293	0 5019
2.5	0.2985	0 4024	0 4369	0.4485	0.4352

\*\*FUSELAGE\*\*

	-----TOP-----
XLOC	CP
38.2	-0 3236
43.4	-0 2797
50 4	-0.2095
56 1	-0 1568
62.5	-0 1744
69 7	-0 2271
76 9	-0.2358
83.4	-0.2007
89 4	-0.2709
95.4	-0.2709
101.1	-0.2095
	---BOTTOM---
38.2	0.1154
43.4	0.1242
50.4	0.0100
56 1	0.1154
62.5	0.1593
69.7	0.1944
76.9	0.2295
83.4	0 2559
89.4	0 2822
95.4	0.1593
101.1	-0.3500

\*\*MISC.\*\*

NO.	CP
1	-0.6320
2	0.3018
3	-1.0656
4	0.3018
5	-1 2990
6	0.1017
7	-0.3629
8	-0.3557
9	-0.3629
10	-0.3557
11	-0 3701
12	-0.3557

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TABLE A45 RUN 290, BC1W6V, DELF=45, CMU=0.5, (BN/B)W=0 25

RUN SEQ	PROPLULSIVE										WING / CANARD		PRESSURES		PAGE 716	
290 1	CLAERO =	0 4167	CDAERO =	0.4210	DCLC =	0 2138	DWLC =	0 4139	BASEPR =		8.1904					
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN					
0.05	2133 75	2119.97	13 79	108.27	-0 05	2 65	0.261	0 478	0 739	86.5494	0.679618E+06					
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR						
0 05	1.04	0 03	-0 06	-0 01	-0.00	0.02	0.62	0.05	0.62	0 33						

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**			
BP=3.375 BP=10.125			BP = 2		BP = 6		BP =12		BP =16		BP = 22		-----TOP----	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0 0	0.4845	-0.3524	0.0	-0.7215	-0.6216	-0.7464	0.3519	0.1272	38 2	-0.3101	1	-0.2981		
2.5	-0 4771	-1 0825	2.5	-0 2971	0 7763	-0.3970	-0.5967	-0 9212	43.4	-0.3101	2	-0 6590		
5.0	-0 4593	-0 9045	5.0	0 6265	0 5766	-0 3471	-0.4968	-0.7714	50 4	-0.1580	3	-0 5146		
10.0	0.0571	-0.7620	10 0	-0.3221	0.3269	-0.2721	-0 4469	-0.5967	56 1	-0 0441	4	-0 2259		
15 0	-0 5127	-0 7264	15 0	0.2770	0 1772	-0.3221	-0 4469	-0.4968	62 5	0 0129	5	-0 4425		
25 0	-0 5127	-0.6730	24 0	0.0274	-0 0475	-0.4219	-0.4469	-0 4718	69.7	-0 0061	6	-0 1538		
35 0	-0 5483	-0.6373	33 0	-0.1538	-0 1538	-0 4425	-0 5868	-0 4425	76 9	-0 0441	7	-0 3910		
50.0	-0.7086		54.0	-0 4425	-0 4425	-0 5868	-0 8755	-0 5146	83 4	-0.0820	8	-0 3601		
56 0	-0 8333	-0 9757	65 0	-0.5951	-0.9561	-0.5951	-0 6711	-0 6141	89 4	-0 1961	9	-0 3446		
65.0	-1 1894	-1.4744	78.5	-0 6141	-2.2481	-0.9561	-1 0511	-1 1271	95.4	-0.2531	10	-0 3601		
76 0	-0 3347	-3.4154	79.5-22.8186	-27 8581	-0.8501	-0.9915	-1 2101	101.1	-0 2151	11	-0 3755			
79.0	-5 1605	-5 8728	80.5-22 1375	-1 4929	-0 8501	-0.9787	-0 3360	---	---BOTTOM---	12	-0.3601			
80.5	-8.5370	-19 2195	81 3-22 5234	115.4856	-0 8630	-1 0044	-1.0944	38 2	0 1459					
81 0	-5 5035	-11.2882	82.0-22.1759	-2.6627	-0 8758	-0 9530	-1 1458	43 4	0 2029					
82 0	-4.1281	-7.8175	84.0-21 8031	-1 5443	-0.8758	-0.9915	-1.1330	50.4	0.0509					
84 0	-0 7473		87.0-11.3136	-1 4929	-0.9273	-1.0173	-0.3745	56.1	0.0319					
87.0	-1.5571		89 0	-1 8912	-1.4581	-1.0251	-1 1334	62.5	-0.3101					
89.0	-1.5641	-2.6090	93 0	2.0217	-1.1179	-0 9787	-0 3910	69 7	-0.4241					
93 0	-0.9561		96 0	0 3514	-0.9478	-0 9787	-1 1024	76.9	-0.3861					
96 0	-0 3481	0 6779	100.0	-0.9787	-1 0714	-0.8859	-0 9478	83.4	-0.2721					
100 0	0 3169	-0.2911	96 0	0.1348	0 0111	0.0885	0 3514	89.4	-0.0630					
96 0	0 6209	0.5829	84.0	0 1194	0 0111	-0 3601	0 6762	95.4	-0.0441					
84 0	0 3169	-0 2911	73 5	-0 1538	-0.3703	0.0627	0 3514	101 1	-0 3481					
73 0	0 6209	0.5829	54 0	-0 5146	-0.4425	-0 4425	-0.1538							
50 0	0 7919	0.6019	33 0	-0 5868	-0 3703	-0 6590	-0 4425							
35.0	0 6626	0.5914	24.0	-0 5717	-0 6216	-0 5967	-0 3720							
25 0	0 3064	-1.1182	10 0	-0.6466	-0 5967	-0.6466	-0 3471							
10 0	0 2174	0 1818	5 0	-0.6466	-0 5717	-0 6466	-0 2472							
5 0	0 0571	0 1640	2 5	-0 5717	-0.5468	-0 6466	-0 2222							
2 5	0.1105	0 3598												

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TABLE A45. (Continued)

RUN SEQ			PROPLULSIVE							WING / CANARD PRESSURES				PAGE 718
290 3	CLAERO =	0 6265	CDAERO =	0 4831	DCLC =	0.2276	DWLC =	0.4383	BASEPR =	8.1904				
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN			
4.08	2133.75	2120.08	13 67	107 82	-0 05	2 66	0.265	0 487	0 753	87.0614	0.676836E+06			
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR				
0 05	1.29	0.13	-0 03	0 00	-0 00	0.02	0.87	0.09	0.84	0.40				

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **	
	BP=3.375	BP=10 125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO. CP	
0.0	-1 0809	-3 2713	0 0	-0.7348	-0 6341	-0 9109	-0.5334	-2.1191	38.2	-0.3966	1 -0.3079	
2 5	-1 2065	-2.1222	2 5	-0.3069	0.5992	-0.9864	-0.6844	-1.6156	43.4	-0.3966	2 -0.5990	
5 0	-0 8116	-1 5118	5 0	0 5489	0 5237	-1.1627	-0 6341	-1.1627	50.4	-0.2242	3 -0.5262	
10 0	0.2657	-1.2065	10 0	-0.3573	0.2972	-1.1123	-0.6089	-0 8606	56.1	-0.0901	4 -0.0896	
15 0	-0.7756	-1.1168	15 0	0 1965	0.0958	-1 0368	-0.6089	-0.7096	62.5	-0.0326	5 -0.4534	
25 0	-0.6679	-0.8475	24 0	0.0203	-0.1055	-0 9361	-0.6089	-0.5586	69.7	-0 0710	6 -0.0896	
35 0	-0 6679	-0 8295	33 0	-0 0896	-0 1624	-0 6717	-0 6717	-0.4534	76.9	-0.1093	7 -0.3859	
50 0	-0.8116		54 0	-0.3807	-0.3807	-0 5262	-0.6717	-0 5262	83.4	-0.1475	8 -0.3859	
56 0	-0.9552	-1.1168	65 0	-0.6456	-1 0479	-0.6073	-0.9139	-0 7223	89.4	-0.2242	9 -0.3859	
65 0	-1 2963	-1.6375	78.5	-0.4924	-2.6380	-0.9904	-1.1437	-1.4694	95.4	-0.2816	10 -0.3548	
76 0	-0 3268	-3.7023	79 5	-22.9757	-31.4524	-0.9034	-0.9682	-1.3181	101.1	-0.2433	11 -0.3704	
79 0	-5 3720	-6.2518	80.5	-22.1596	-1 4218	-1.0201	-0.9812	-0.3201	---	---	12 -0.3704	
80 5	-8 6800	-20.0468	81 3	-22 4185	112 0886	-0 9682	-1 0330	-1.5255	38 2	0.1590		
81 0	-5.6341	-11 8554	82 0	-22 5355	-2 8216	-0 9812	-1.0070	-1 6292	43.4	0.2356		
82 0	-4 2343	-8 3041	84 0	-21.9782	-1 9143	-0.8256	-1.0459	-1.5903	50 4	0.0632		
84 0	-0 7349		87.0	-12 5033	-1 7458	-1.0978	-1.0070	-0.3201	56.1	0.0440		
87 0	-1 6162		89 0	-3 0213	-1.5554	-1.0720	-1.2436	-1.6178	62.5	-0.2242		
89 0	-1 6418	-2 9253	93 0	2 0779	-1.1657	-1.0409	-0.4171	-0.3548	69.7	-0.3774		
93 0	-1 0479		96 0	0 2846	-1 0565	-1 0565	-1.0876	-1.4463	76.9	-0.3583		
96.0	-0 3966	0 7337	100 0	-0 8382	-1.0720	-0.9785	-0.9629	-0.3548	83 4	-0 1858		
100 0	0.4655	-0 3391	96 0	0 2378	0.0975	0.1286	0.4093	0 3158	89 4	0.0057		
96 0	0.6379	0 5421	84 0	0 2222	0 1754	-0 3704	0 7992	0 6120	95.4	-0.0135		
84 0	0.4655	-0 3391	73.5	-0 0168	-0.3807	0 2742	0.5653	0.4925	101.1	-0 3966		
73 0	0 6379	0 5421	54.0	-0.3807	-0 2351	-0.1624	-0.0896	0 1287				
50 0	0 7720	0 6187	33 0	-0 5990	-0 3807	-0 4534	-0.2351	-0.0168				
35 0	0 7505	0 6069	24 0	-0.5334	-0 6089	-0 6341	-0 2817	-0 0552				
25 0	0.3735	-1.2245	10.0	-0 6844	-0.5082	-0.6592	-0 3824	0 2015				
10 0	0.3376	0.2837	5.0	-0.5837	-0.5082	-0 6341	-0.3320	0 2015				
5.0	0 1939	0.2657	2.5	-0 6341	-0.5082	-0 6844	-0.3069	0.3470				
2 5	0.4094	0 5530										

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TABLE A45 (Concluded)

RUN SEQ	290 5	CLAERO =	1.3032	CDAERO =	0 8575	DCLC =	0 2567	DWLC =	0 4893	BASEPR =	8.1904	PAGE	720
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
12 15	2133 68	2120 46	13 22	106 02	-0 04	2 72	0 279	0 514	0 793	88.9776	0.665757E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
0 04	2 05	0 59	-0.03	-0 00	-0.00	0.01	1 68	0.09	1 54	0.80			

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**		
%X/C	BP=3.375 CP	BP=10.125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XLOC	CP	NO.	CP	
0 0	-8 0059	-3 0120	0 0	-1 2139	-0.7976	-0.7716	-1 9427	-4.1288	38.2	-0.5866	1	-0.9581	
2 5	-5.2213	-3 0863	2.5	-0 5633	0.1654	-0.9277	-1.4742	-3.6603	43.4	-0.5074	2	0.0200	
5 0	-1.6196	-3 0863	5 0	0.4256	0.1394	-1 0318	-1.1879	-3.7124	50.4	-0.3290	3	-0 8076	
10.0	0.4226	-3 2534	10 0	-0 6414	-0.0949	-1.1619	-0.9277	-3 0878	56.1	-0.1903	4	0.1705	
15 0	-1 4710	-3 4391	15 0	-0 0168	-0 2771	-1.2921	-0.8756	-2 0207	62.5	-0.1112	5	-1.3343	
25 0	-1 1555	-4 1631	24 0	-0 1730	-0 4853	-1 2921	-0 8496	-0.9277	69.7	-0.1508	6	0.0200	
35 0	-1 0626	-3 2348	33 0	-0 2809	-0 4314	-1 2591	-0 8829	-0.7324	76.9	-0 1705	7	-0 6304	
50 0	-1.1184		54 0	-0.5067	-0 6572	-1.1839	-0.5819	-0 8076	83.4	-0.1705	8	-0.6143	
56 0	-1 2483	-1 0626	65.0	-0 8441	-1 2006	-1.2799	-1.1214	-0 9828	89.4	-0.3092	9	-0.6143	
65 0	-1 6010	-1 3225	78 5	-0 3092	-3 8154	-1 5968	-1 4384	-1.5374	95.4	-0.3489	10	-0.5981	
76 0	-0 5428	-3 6062	79.5-23	8491	-46 0428	-1.4276	-1.2936	-1 3740	101.1	-0.2894	11	-0.5820	
79 0	-5.9640	-6 3539	80.5-22	9106	-5 8904	-1.7090	-1.2266	-0 5029	---BOTTOM---			12	-0 5981
80 5	-9 3618	-21 8255	81.3-23	3530	107 8493	-1 5616	-1.2668	-1.4008	38 2	0.2454			
81 0	-6 1183	-13 0203	82 0-23	6078	-5 2606	-1 5482	-1.2936	-1 4544	43.4	0.2851			
82 0	-4 7244	-9 4419	84.0-23	0316	-3 7461	-1 4678	-1.2936	-1 4142	50.4	0 0870			
84 0	-1 0390		87 0-16	0224	-2 9822	-1 8029	-1.3338	-0 5430	56.1	0.1068			
87 0	-1 9235		89 0	-6 2900	-2.5170	-1 9526	-1 5979	-1.5172	62.5	-0.0715			
89 0	-1 9335	-3 6965	93.0	1.7399	-1 6462	-1.7269	-0.6626	-0 6143	69 7	-0.1310			
93 0	-1.2601		96 0	0 1597	-1.0657	-1 5818	-1.4043	-1 5979	76.9	-0.0319			
96 0	-0.5866	0 8199	100 0	-0 0821	-0.5820	-1 7430	-1 5011	-0.6143	83.4	0.1860			
100 0	0 5822	-0 4677	96 0	0 6596	0 4177	0 0630	0 1597	0 1920	89.4	0 2454			
96 0	0 6812	0 4831	84 0	0 3210	0 4177	-0 6465	0.4661	0 6596	95.4	0 0870			
84 0	0 5822	-0 4677	73 5	0.2457	-0 5067	0 2457	0.3210	0 4715	101.1	-0.6460			
73 0	0 6812	0 4831	54 0	0 1705	0.1705	0.1705	0 1705	0 1705					
50 0	0 8397	0 6614	33 0	-0 1305	-0 5067	0 0200	0 1705	0 1705					
35 0	0 8125	0 5897	24 0	-0 2771	-0 3291	-0 0428	0 1654	0 1654					
25 0	0 5155	-1 5825	10.0	-0 8236	-0 6414	-0 3291	0 1654	0 3962					
10 0	0 4783	0 3855	5 0	-1 0578	-0 7195	-0 4593	0.1914	0.3962					
5 0	0 4412	0.4226	2 5	-1 0578	-0 7455	-0 4593	0 2174	0 3210					
2 5	0 6083	0 4412											

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TABLE A46 RUN 291, BC1W6V, DELF=45, CMU=1.0, (BN/B)W=0.25

RUN SEQ				PROPLUSIVE		WING / CANARD		PRESSURES			PAGE 721
291 1	CLAERD =	0.2638	CDAERD =	0.5784	DCLC =	0.4660	DWLC =	0.9072	BASEPR =	8 1904	
ALPHA	PTOT	PSTAT	0	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0 08	2133.96	2127.52	6 44	73 91	-0.05	2.66	0.569	1.048	1.618	86.9237	0.465251E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0 05	1.64	-0.27	-0.24	-0 01	0.00	0.01	0.71	0.01	0.71	0.37	

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	0 4754	-0 3630
2.5	-0 4773	-1.2014
5 0	-0.4773	-0.9346
10.0	0.4754	-0 8584
15.0	-0.5535	-0.8204
25 0	-0.5917	-0 7822
35 0	-0 6298	-0.7822
50.0	-0.7822	
55 0	-0.9728	-1.1633
65.0	-1 4301	-1 7731
76 0	-0 3630	-4 2501
79.0	-6.4223	-7.4132
80.5	-13.1310	-32 5251
81.0	-8 0692	-18 4683
82 0	-5 8135	-12 3057
84 0	-0 8067	
87.0	-1.5769	
89.0	-2.0407	-3.5451
93 0	-1 1868	
96 0	-0.1702	0.6836
100.0	0.9683	-0.9022
96 0	0 5617	0.5617
84 0	0 9683	-0 9022
73 0	0.5617	0.5617
50.0	0 7650	0.6023
35 0	0 6659	0 5516
25 0	0.2848	-1 3158
10 0	0.2087	0 1705
5 0	0 0943	0 0943
2 5	0 0943	0.3992

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0 9677	-0.8609	-0.3266	0.8486	0.0474
2.5	-0.3266	1.0089	0.5282	-0.6471	-0.6471
5.0	0.7418	0.7952	0 1008	-0.4868	-0.7005
10.0	-0.3266	0 3679	-0.1129	-0.3800	-0 6471
15.0	0 3679	0.2076	-0.2732	-0.3266	-0.5403
24.0	0 2076	-0.0061	-0.3266	-0.3266	-0 4868
33.0	-0.0722	-0.2266	-0.3810	-0.5355	-0.5355
54.0	-0.3810	-0.3810	-0.5355	-1.4622	-0.5355
65.0	-0.7395	-1 4713	-0.7395	-0.6988	-0 6988
78.5	-1.5934	-3 3011	-1.1462	-1.2275	-1.3494
79 5-50	8195	-60.7227	-1.0268	-1.0543	-1.3845
80.5-48	9755	-1.9896	-1.1369	-0.9717	-0.3115
81.3-48	3702	230 0229	-1 0543	-1 0268	-1.2744
82.0-47	5723	-4.4105	-1.0268	-0 9993	-1.3294
84 0-45	8121	-2.4022	-0 8893	-1 0268	-1.3294
87.0-13	9014	-1.8796	-1.1094	-0.9442	-0.3115
89.0 2	2998	-1.5727	-1.1755	-1.2418	-1.5064
93.0 3	1603	-1 2749	-1.2086	-0 4143	-0.3812
96.0 0	1484	-1.0762	-1.1424	-1.2086	-1 6058
100.0 -1	9367	-1.1094	-0 9770	-1.1755	-0.3481
96 0 0	0160	0 0160	0 2146	0 3470	0.2477
84.0 0	1153	0 4132	-0 4143	0.7442	0.6780
73 5 -0	2266	-0 3810	0.2367	0 5456	0 5456
54 0 -0	3810	-0.3810	-0.3810	-0.2266	0.0822
33 0 -0	6900	-0 3810	-0 6900	-0.2266	-0.2266
24.0 -0	8609	-0 9677	-0.5938	-0.2732	-0 1129
10 0 -0	7539	-0.8609	-0 7539	-0 2198	-0 0722
5.0 -0	7005	-0 8073	-0.9143	-0 1663	0 0822
2 5 -0	6471	-0 7539	-1 0211	-0 2198	0 2367

\*\*FUSELAGE\*\*

	-----TOP----
XLOC	CP
38.2	-0.3329
43.4	-0.3329
50.4	-0.1702
56.1	-0.0483
62.5	0.0737
69.7	-0.0076
76.9	-0.0483
83.4	-0.1296
89.4	-0.1702
95.4	-0.2516
101.1	-0.2109
	---BOTTOM---
38.2	0.1550
43.4	0.1957
50.4	-0.0483
56.1	-0.1702
62.5	-0.6175
69.7	-0.6988
76.9	-0.5768
83.4	-0.3735
89.4	-0.1702
95.4	-0.1296
101.1	-0.4955

\*\*MISC \*\*

NO.	CP
1	-0.2266
2	-0.6900
3	-0.5355
4	-0.2266
5	-0.3810
6	-0 2266
7	-0.3481
8	-0.3481
9	-0.3150
10	-0.3481
11	-0.3481
12	-0.3481

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TABLE A46 (Continued)

PAGE 723

RUN SEQ	PROPLULSIVE WING / CANARD PRESSURES										BASEPR = 8.1904	
291 3	CLAERO = 0 3103	CDAERO = 0 5364	DCLC = 0.4821	DWLC = 0 9278	HGT		RN					
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	87 6989 0	465629E+06	
4.05	2134 11	2127.66	6 44	73.89	-0 05	2 68	0.562	1.032	1 594	CLTR	CDTR	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	0.74	0.38		
0 05	1 72	-0.20	-0 09	-0.01	0 00	-0.00	0.77	0 16				

***** CANARD *****				***** WING *****						**FUSELAGE**		**MISC.**				
BP=3 375		BP=10.125		BP = 2		BP = 6		BP =12		BP =16		BP = 22		----TOP----		
%X/C	CP		CP	%X/C	CP		CP		CP		CP		CP	NO.	CP	
0 0	-0 9501		-3 7321	0 0	-0 9298		-0 8228		-0 9298		-0 2887		-1.7844	38 2	-0.4297	1 -0.2421
2 5	-1 4075		-3 2367	2.5	-0 3955		0.8331		-0 7694		-0.6626		-1 8914	43.4	-0 4297	2 -0.8599
5 0	-0 9501		-2 0553	5 0	0 0.6195		0 7263		-0 5023		-0 5558		-1 0366	50.4	-0.2264	3 -0.7055
10 0	0.8029		-1 2932	10.0	-0 3421		0 4058		-0.6093		-0 6093		-0.7694	56 1	-0 0638	4 -0 2421
15 0	-0.7977		-1 1407	15.0	0 2455		0 1921		-0 6626		-0 5023		-0.7694	62 5	0 0175	5 -0.5510
25 0	-0 7215		-0.9501	24.0	0 0319		-0.0750		-0.6093		-0 5023		-0 5558	69.7	-0.0231	6 -0.2421
35 0	-0.7215		-0.9121	33.0	-0 2421		-0 2421		-0.7055		-0 7055		-0.5510	76.9	-0.0638	7 -0 3967
50 0	-0 9121			54.0	-0 3965		-0.3965		-0 7055		-1.1688		-0.5510	83.4	-0.1044	8 -0.3967
56 0	-0 9883		-1 3313	65 0	-0 7143		-1 4462		-0.6330		-0.8363		-0.7550	89 4	-0.1857	9 -0 3967
65 0	-1.4456		-1 9029	78.5	-1.5275		-3 3573		-1.0803		-1.1617		-1 2430	95 4	-0.2264	10 -0 3636
76 0	-0 3023		-4 5705	79 5-49	2114		-61.8667		-0.9322		-1 0148		-1.3174	101.1	-0.2264	11 -0.3967
79.0	-6 6284		-7.7716	80.5-48	4695		-2.4728		-0.9872		-0 9872		0 4096	---BOTTOM---		12 -0 3967
80 5	-13 2566		-33 0917	81 3-48	8265		219 3437		-0.9597		-1.0698		-1 2624	38.2	0.1802	
81.0	-8.1398		-18 9230	82 0-48	6340		-4 5360		-0 9597		-1.0148		-1 2899	43.4	0.2209	
82 0	-5 9666		-12 7064	84 0-46	7636		-2 4452		-0 9322		-1 0148		-1 2073	50.4	0.0989	
84 0	-0 8773			87 0-15	5951		-1.9500		-1.0423		-1 0698		-0.3546	56.1	-0.0638	
87 0	-1 6475			89 0 2	9463		-1 4228		-1.1579		-1 2241		-1 4228	62 5	-0.5110	
89 0	-2 0969		-3 8046	93 0 3.	7406		-1 2571		-1 0917		-0.4297		-0.4297	69.7	-0 6736	
93 0	-1 2023			96 0 0	3646		-1 0917		-1 1249		-1 1579		-1 4228	76.9	-0 6330	
96 0	-0 2671		0 7495	100 0 -1	6874		-1 2571		-1 0586		-0 9925		-0.3967	83 4	-0 4704	
100 0	1 2374		-0 9583	96.0	-0 0988		-0 1319		0 0667		0.3315		0 2653	89 4	-0 2264	
96 0	0.5868		0 5462	84 0	0.0005		0 0005		-0 3967		0 6625		0.6294	95.4	-0.1857	
84 0	1 2374		-0 9583	73 5	-0 2421		-0 3965		-0.2421		0 5301		0.5301	101 1	-0.5517	
73 0	0 5868		0 5462	54 0	-0 7055		-0 7055		-0 7055		-0.2421		0 0667			
50 0	0 8308		0 6681	33.0	-0 7055		-0 3965		-0 8599		-0 5510		-0 2421			
35 0	0 6886		0 5742	24 0	-0 7694		-0 8228		-0 8228		-0 5023		-0 0750			
25 0	0 3455		-1 4075	10 0	-0 6093		-0 7160		-0 8228		-0.5023		0 0667			
10 0	0 3074		0 2312	5 0	-0 7160		-0 7694		-0 7694		-0 6093		0 2212			
5 0	0 2312		0 2693	2 5	-0 7160		-0.7694		-0 8228		-0.5558		0.3757			
2 5	0 4217		0 5361													

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TABLE A46. (Concluded)

RUN SEQ			PROPLULSIVE	WING / CANARD	PRESSURES		PAGE 725				
291 5	CLAERD =	0 9046	CDAERD =	0 8052	DCLC =	0 5439	DWLC =	1.0293	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11 95	2134.25	2128.14	6.10	71.91	-0.04	2.72	0.591	1.083	1.674	88 8088	0.453373E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0 04	2 48	0.24	-0.05	-0 02	0 00	-0.00	1.49	0 21	1.37	0.70	

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-6 8032	-3 1433
2 5	-7 0042	-3 1834
5 0	-2.6607	-3 2237
10 0	0.8786	-3 4247
15 0	-1.3736	-3.7063
25 0	-1.2128	-4.3097
35 0	-1.1323	-3.4650
50 0	-1.2932	
56 0	-1.3335	-1.2932
65 0	-1.7758	-1.5345
76 0	-0.5291	-4 1889
79 0	-7.3259	-7 5673
80 5	-14 5041	-34 5369
81 0	-9.1041	-20.3981
82 0	-6 6073	-14.3008
84 0	-1.1492	
87 0	-1.9330	
89 0	-2 4766	-4 9226
93 0	-1 5325	
96 0	-0 5027	0.8275
100 0	1.2567	-1 1035
96 0	0 6559	0.4843
84 0	1 2567	-1.1035
73 0	0 6559	0 4843
50 0	0 7846	0.6988
35 0	0 8383	0.5970
25 0	0.5166	-1 6552
10 0	0.5166	0 3557
5 0	0 4362	0 3960
2 5	0 5970	0.4362

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0 0	-0 8612	-0.5793	-1.2558	-1 1995	-4.3567
2.5	-0 5793	0.3791	-1.1431	-1.0868	-3 2855
5 0	0.4918	0 3791	-1.3122	-0.9176	-2 2707
10 0	-0.6357	-0 0156	-1 4250	-0 8612	-1.3122
15 0	0.1535	-0.1847	-1.4814	-0.8049	-1.0304
24 0	-0.1283	-0.3539	-1.4250	-0 7485	-0.8049
33 0	-0.3047	-0.4677	-1.2827	-0.7937	-0.7937
54 0	-0.4677	-0 6307	-1.1197	-1 2827	-0 7937
65 0	-0.8889	-1.5754	-1.1035	-1 2322	-0 8889
78 5	-1 1464	-4.0644	-1 6183	-1.6183	-1.3610
79 5	-51.3457	-76.2274	-1.7008	-1 3815	-1.2364
80 5	-49.7207	-5.0976	-1.7298	-1.4105	-0 4524
81 3	-50.9984	220.6077	-1 3815	-1.3815	-1.2654
82 0	-51.0558	-6.3750	-1.5556	-1.2654	-1.2072
84 0	-49.5176	-3 7912	-1.2654	-1.3523	-1.3523
87 0	-21 7905	-2 9202	-1 4976	-1 2364	-0.3363
89 0	0 1958	-2.5635	-1.8300	-1 5856	-1.4808
93 0	4 2476	-1 5156	-1.5856	-0 6774	-0.5726
96 0	0 3705	-1.2712	-1 4458	-1 2712	-1.4458
100 0	-1 6554	-1.4110	-1.1664	-1 2013	-0 6075
96 0	0 1958	0.0561	0 0910	0 4054	0 3355
84 0	0 1958	0.1260	-0.6075	0 7547	0 6848
73 5	-0.1417	-0.4677	0 3473	0 6733	0 6733
54 0	-0.4677	-0 4677	-0.3047	0.0212	0.1843
33 0	-0.6307	-0.4677	-0.6307	-0 3047	0 1843
24 0	-0.5230	-0.6922	-0.6922	-0.2411	0.0408
10 0	-0.5230	-0.6922	-0 9176	-0 4102	0 5103
5 0	-0.6357	-0 6357	-0 8612	-0 4666	0 5103
2 5	-0.5230	-0.6922	-0.9176	-0.4102	0.5103

\*\*FUSELAGE\*\*

----	TOP	----
XLOC	CP	
38 2	-0.6314	
43.4	-0.5456	
50.4	-0.2882	
56.1	-0.1165	
62.5	-0.0307	
69.7	-0.0736	
76 9	-0.1594	
83.4	-0.1594	
89.4	-0.2452	
95.4	-0.3311	
101.1	-0.2882	
----	BOTTOM	----
38 2	0.2697	
43.4	0.3126	
50.4	0 0552	
56.1	0.0122	
62.5	-0.3311	
69.7	-0.5027	
76 9	-0.5027	
83.4	-0 2882	
89.4	0.0122	
95.4	-0 0307	
101.1	-0.7172	

\*\*MISC.\*\*

NO.	CP
1	-0 7937
2	-0.7937
3	-0.7937
4	0.0212
5	-0.7937
6	0.0212
7	-0.6075
8	-0 6075
9	-0 6075
10	-0.6075
11	-0.6425
12	-0 6425

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TABLE A47 RUN 292, BC1W6V, DELF=45, CMU=2 0, (BN/B)W=0.25

RUN SEQ	292 1	CLAERO = -0.0390	CDAERO = 0 7616	DCLC = 0 8912	DWLC = 1.7183	BASEPR = 8.1904	PAGE 726
ALPHA	PTOT	PSTAT	Q VEL	YAW H/C	CMUC	CMUW	CMUT
-0 08	2134 46	2131.07	3.39 53.58	-0 05 2 67	1 089	1.986 3 075	HGT RN
BETA	CL	CD	CM	CROLL	CN	CY	CNTR
0 05	2.57	-0 86	-0 47	-0 01	-0.00	-0.03	0.78 0 01
							CLTR CDTR
							0.78 0.38

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3 375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO CP	
0 0	0 2061	-1 6037	0 0	-1.4330	-1.2300	-0 6211	1.1039	0.2921	38.2	-0.5423	1 -0.1710	
2 5	-0 8797	-2 2550	2.5	-0 4182	1 3069	0 4950	-0.4182	-0 9256	43 4	-0 5423	2 -1.3445	
5 0	-0 8073	-1 3864	5 0	0.7995	0 9010	0 1906	-0 3167	-0 9256	50.4	-0 2334	3 -0.4644	
10 0	1 0024	-1 1693	10 0	-0 4182	0.3936	-0 4182	-0.3167	-0.7226	56.1	-0.0789	4 -0.1710	
15 0	-0 7350	-1.0969	15.0	0 3936	0.1906	-0.3167	-0.4182	-0 7226	62.5	-0.0017	5 -0.4644	
25 0	-0.7350	-0 9521	24.0	0 0892	-0 1138	-0.4182	-0.3167	-0.6211	69.7	-0 0017	6 -0.1710	
35 0	-0.8073	-1.0969	33 0	0 1224	-0.1710	-0 4644	-0.4644	-0.7578	76.9	-0.0789	7 -0.5065	
50 0	-1 0969		54.0	-0.4644	-0 4644	-0.7578	-2.5181	-0.7578	83.4	-0.1561	8 -0.5065	
56 0	-1.2416	-1 6037	65.0	-0.8513	-2.2415	-0.7740	-0.7740	-0.9284	89.4	-0.2334	9 -0.3807	
65.0	-1.7484	-2 3276	78.5	-3.3227	-4.1724	-1.3919	-1 3919	-2 8595	95.4	-0.3878	10 -0.5065	
76.0	-0.5177	-5 6573	79 5	-94.9906	-112 1830	-1.0877	-1.2968	-3.5438	101.1	-0.2334	11 -0.4436	
79 0	-8 3357	-9 9284	80.5	-92.0126	-3 2304	-1.1923	-1.4536	-0.3040	---BOTTOM---		12 -0.5065	
80 5	-20 5266	-54.1780	81.3	-93.5282	424.2244	-1.1400	-1.2968	-1 7671	38.2	0.1528		
81 0	-12 3226	-30 2454	82 0	-93 1095	-7.0970	-1.1400	-1.1923	-3.0736	43.4	0.1528		
82 0	-8 5081	-19.6383	84.0	-88.7192	-3.3870	-1 0877	-1 2445	-2 6032	50 4	-0 1561		
84 0	-0 8265		87 0	-22.1987	-2 4464	-1.2968	-1.1923	-0 4085	56 1	-0.4651		
87 0	-1.4013		89 0	8 5466	-2 3925	-1.5124	-1.7009	-2.4552	62 5	-1.0829		
89 0	-2 7050	-4.8677	93.0	5 8435	-1.8267	-1.5124	-0 5065	-0 5065	69 7	-1.1602		
93 0	-1.3919		96.0	0 3109	-1 5124	-1.4495	-1 6381	-1.8896	76.9	-0.7740		
96 0	0 1528	0.6935	100 0	-3 7128	-1 6381	-1 3237	-1 7009	-0 4436	83.4	-0 3106		
100 0	2 0838	-1 9326	96 0	-0 2550	0 1851	0 2481	0 2481	0 3109	89 4	-0.0789		
96 0	0 4618	0 4618	84 0	0 2481	0 5623	-0 5065	0.6881	0.5623	95.4	-0.3106		
84 0	2 0838	-1.9326	73 5	0 1224	-0.4644	0 4157	0 4157	0 4157	101.1	-0.8513		
73 0	0 4618	0 4618	54.0	-0.4644	-0 4644	-0.1710	0.1224	0.1224				
50 0	0.7707	0 6163	33 0	-1 3445	-0 4644	-0 4644	-0.7578	-0.4644				
35 0	0.6404	0 4957	24.0	-1.3315	-1.5344	-1.0271	-0.4182	-0 1138				
25 0	0 2061	-1 6037	10.0	-1 1285	-1.2300	-1.4330	-0 4182	-0.1710				
10 0	0 1337	0 1337	5 0	-1 1285	-1 3315	-1 7374	-0 3167	0 1224				
5.0	0.0613	0 0613	2.5	-1.1285	-1 3315	-1.7374	-0 3167	0 1224				
2 5	0.1337	0 2785										

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TABLE A47 (Continued)

RUN SEQ	292 3	CLAERO = -0 0995	CDAERO = 0.7578	DCLC = 0 9651	DWLC = 1.8406	BASEPR = 8.1904	PAGE 728				
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.16	2134 46	2131 07	3 39	53 57	-0 05	2.67	1.124	2.045	3.169	87.4664	0.338192E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR
0.05	2.71	-0.71	-0.32	-0 03	0 01	-0 02	0 86	0.16		0.83	0.40

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC.**	
	BP=3 375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO	CP
0 0	-1.8932	-3.7031	0.0	-1.3315	-0.9256	-1.4330	-0 0123	-1.7374	38.2	-0.6196	1	-0 4644
2.5	-2 1828	-3 1962	2.5	-0.5197	0 7995	-0.1138	-0.7226	-1.9403	43.4	-0.5423	2	-1.3445
5.0	-1.0969	-2 8342	5 0	0.6980	0 7995	-0 1138	-0 8241	-1.3315	50.4	-0.3106	3	-0.7578
10 0	1 1472	-2.8342	10 0	-0.4182	0.4950	-0.4182	-0.6211	-1.1285	56.1	-0.1561	4	-0 4644
15 0	-1.0969	-1 8208	15.0	0 3936	0.2921	-0 5197	-0.6211	-0 9256	62.5	-0.1561	5	-0.7578
25 0	-0 9521	-1.1693	24.0	0.0892	-0 0123	-0.5197	-0.5197	-0 7226	69 7	-0 0789	6	-0.1710
35 0	-0.8797	-1 0969	33 0	-0 1710	-0.1710	-0 7578	-0 7578	-0 7578	76.9	-0.1561	7	-0.3807
50 0	-1.1693		54 0	-0 4644	-0.7578	-0.7578	-2.2247	-0.7578	83.4	-0.2334	8	-0.4436
56.0	-1 2416	-1.6037	65.0	-0.8513	-2 3960	-0.9285	-0.9285	-0.9285	89 4	-0.3106	9	-0.3807
65.0	-1.8208	-2 3276	78 5	-3.4002	-4.4042	-1.3147	-1.3920	-1.6236	95.4	-0.3878	10	-0.4436
76.0	-0.4454	-5 4403	79.5-93	5783-112	8612	-1.1400	-1.2968	-1.6626	101.1	-0.3878	11	-0.5065
79 0	-8.4082	-10.0007	80.5-91	4377	-3 7529	-1 2445	-1.1400	-0.6175	---	---BOTTOM---	12	-0.4436
80.5	-20.7357	-53 3938	81.3-92	3252	400.4893	-1 2968	-1.4013	-1 6626	38.2	0.1528		
81.0	-12.2182	-30 2454	82.0-92	2721	-7.2017	-1.1923	-1.2445	-1.5058	43.4	0.0756		
82.0	-8.5603	-19 8473	84.0-88	7723	-3.7007	-1.1923	-1.4013	-1.6626	50.4	-0.0789		
84.0	-0 8265		87.0-24	6548	-3 0214	-1 2968	-1.2968	-0.4607	56.1	-0.4651		
87.0	-1.5058		89.0	8.8612	-2.2040	-1 3237	-1.5752	-1.7639	62.5	-1.0058		
89.0	-2.7822	-5.1767	93.0	6.5977	-1.7010	-1.2609	-0.4436	-0.3807	69.7	-1.1602		
93 0	-1.4692		96.0	0.4995	-1.3866	-1 1980	-1 3866	-1.8896	76.9	-0.9285		
96.0	0 0756	0.6934	100 0	-2 6440	-1 6381	-1 1980	-1.1980	-0 3807	83.4	-0.7740		
100 0	2 1610	-2 0097	96.0	-0 3179	-0 2550	0 0594	0 3737	0 1851	89.4	-0.4651		
96 0	0 4618	0 3845	84 0	-0 0664	-0 1292	-0 3807	0 8139	0 6881	95.4	-0 4651		
84 0	2.1610	-2 0097	73.5	-0 4644	-0 4644	0 1224	0 4157	0 4157	101.1	-0.9285		
73.0	0 4618	0 3845	54.0	-1 0512	-1.3445	-1.0512	-0 1710	0.1224				
50 0	0.6934	0 5390	33.0	-1 3445	-0 4644	-1.3445	-0.7578	-0 1710				
35 0	0 7129	0 4957	24.0	-1 0271	-1 2300	-1.1285	-0.7226	-0.3167				
25 0	0.2785	-1 6760	10.0	-1.0271	-1 0271	-1 2300	-0.6211	-0 1710				
10 0	0 3508	0.2061	5 0	-0 9256	-1.0271	-1 2300	-0.7226	0 1224				
5 0	0 2061	0 2061	2.5	-1.0271	-0.9256	-1 0271	-0 5197	0.1224				
2.5	0.4233	0.4957										

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TABLE A47 (Concluded)

RUN SEQ				PROPLULSIVE	WING / CANARD	PRESSURES			PAGE 730
292 5	CLAERO =	0 1906	CDAERO =	0 8547	DCLC =	1 0847	DWLC =	2 0372	BASEPR = 8.1904
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	HGT RN
12 13	2134.60	2131 32	3 28	52.67	-0 04	2 74	1.177	2 140	3 318
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR CDTR
0 04	3 31	-0 26	-0 17	-0 02	0 00	-0.04	1.32	0.33	1 22 0.61

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC CP	NO CP	
0.0	-6 7051	-3 4108	0.0	-1 4075	-0 7778	-1.5125	-0 9877	-4.3460	38 2 -0.6712	1 -0.5108	
2 5	-6.5554	-3 4108	2 5	-0 5680	0 4815	-1.5125	-0.8827	-3 1916	43 4 -0 6712	2 -1.4210	
5 0	-6 2559	-3 6356	5 0	0 5865	0 4815	-1 5125	-0 8827	-1 8274	50.4 -0.3517	3 -1.1176	
10 0	1 2309	-3 7102	10.0	-0 5680	0 1667	-1.5125	-0.7778	-1.1976	56 1 -0 1120	4 0 0961	
15 0	-1 3894	-4 1595	15 0	0 2717	0 1667	-1 4075	-0 8827	-0 9877	62.5 -0 1120	5 -0.8142	
25 0	-1 3146	-4 7586	24 0	-0 0432	-0 2531	-1 1976	-0 7778	-0.7778	69.7 -0 0322	6 0.0961	
35.0	-1 1649	-4 1595	33 0	-0 2073	-0 2073	-1 4210	-1.1176	-0 5108	76.9 -0.1120	7 -0 4892	
50 0	-1 4643		54.0	-0 5108	-0 5108	-1.1176	-1.7244	-0.8142	83.4 -0.1120	8 -0 5543	
56 0	-1 5392	-1 8387	65 0	-0 9109	-2.1889	-1 0707	-1.2304	-0.9908	89 4 -0.1919	9 -0 5543	
65 0	-2 0633	-1.8387	78 5	-2 1889	-4 6653	-1 3902	-1.7096	-1.7096	95.4 -0.2718	10 -0 4892	
76 0	-0.4911	-4 9081	79 5	-97 7861	-123 4592	-1.8581	-1 4257	-1.4797	101.1 -0.2718	11 -0.4892	
79.0	-9 1758	-9 3255	80.5	-95.1392	-7.1543	-1.4797	-1.5338	-0.5069	---BOTTOM---	12 -0.5543	
80 5	-22 2867	-55 6857	81 3	-96.9231	404 7815	-1 6959	-1 3716	-1 3716	38.2 0 2873		
81 0	-13 3154	-31 7446	82 0	-96 0022	-8 6138	-1.4257	-1 4257	-1 4257	43.4 0.2873		
82 0	-9 2081	-21 3140	84 0	-92 8155	-4 8846	-1.3716	-1.5338	-1 5878	50.4 -0.0322		
84 0	-1 2635		87.0	-23 8002	-3 5334	-1.6419	-1.4257	-0.3988	56.1 -0.2718		
87 0	-1.9121		89.0	8.6790	-2 5698	-1 5946	-1.5946	-1.5946	62 5 -0.8310		
89 0	-3.0677	-6.0232	93 0	6 4032	-1 8547	-1.5296	-0.6193	-0.4892	69.7 -0.9908		
93 0	-1 7096		96.0	0 4212	-1 7246	-1.3995	-1.3345	-1 6596	76.9 -0 9109		
96 0	-0 1120	0 8466	100 0	-2.5048	-1 9197	-1.1394	-1 2044	-0 4892	83.4 -0.5913		
100 0	2 2045	-2 0292	96 0	-0 2941	-0 1641	0 2911	0.4862	0 3561	89 4 -0 1919		
96 0	0 6068	0 3673	84 0	0 0310	0 0310	-0 5543	0 8113	0 8113	95.4 -0.3517		
84 0	2 2045	-2 0292	73.5	0 0961	-0 5108	0 0961	0 7029	0 7029	101 1 -1.0707		
73 0	0 6068	0 3673	54 0	-0.8142	-0.8142	-0 5108	0.0961	0 0961			
50.0	0 8466	0 6068	33.0	-1 4210	-0 5108	-1.1176	-0 5108	-0 0961			
35 0	0 8565	0.5571	24 0	-1 0927	-1 1976	-1 1976	-0.5680	-0.0432			
25 0	0.4074	-1 9884	10 0	-1 0927	-0 8827	-1 1976	-0.7778	0 3995			
10 0	0 4074	0 4074	5.0	-0 8827	-1 0927	-1 1976	-0.9877	0 3995			
5.0	0 3325	0 3325	2 5	-0 8827	-0 9877	-1.1976	-0 8827	0 7029			
2 5	0 5571	0 2576									

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TABLE A48. RUN 296, BC1W6V, DELF=45, CMU=0 5, (BN/B)W=0 25, (BN/B)C=0.5

RUN SEQ	PROPLULSIVE WING / CANARD PRESSURES										PAGE 731	
296 2	CLAERO =	0 4977	CDAERO =	0 4903	DCLC =	0 2721	DWLC =	0.3917	BASEPR =	8.1905		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
-0.08	2142 17	2128 72	13.45	106 50	-0.05	2 65	0.333	0.453	0 785	86.7669	0.676155E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0 05	1.16	0 07	-0.30	0 01	-0.01	0.00	0 69	-0 19	0 69	0 41		

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**			
BP=3.375 BP=10.125			BP = 2 BP = 6 BP =12 BP =16 BP = 22					----TOP----		NO. CP			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP			
0 0	0 5295	0 1281	0.0	-0 5828	-0 3525	0 1338	0 2361	-0.0966	38.2	-0.2819	1	-0.3282	
2.5	-0 4010	-0 8206	2.5	-0 2757	0.5688	-0.1989	-0.5316	-0.7876	43.4	-0.3014	2	-0.0321	
5 0	-0 3644	-0 5834	5.0	0.5432	0.4152	-0.2246	-0.3781	-0.7108	50.4	-0.1650	3	-0.4761	
10 0	-0.1273	-0.5834	10 0	-0 3269	0 1849	-0.3525	-0.3269	-0.5316	56.1	-0.0676	4	0.1158	
15 0	-0 4557	-0 6017	15 0	0.1849	0.0570	-0 3781	-0.2501	-0.4549	62.5	-0.0481	5	-0.4022	
25.0	-0 4557	-0 5104	24.0	-0 0198	-0 0710	-0 4293	-0 2757	-0 4037	69.7	-0.0676	6	-0.0321	
35.0	-0 4374	-0 4557	33 0	-0.1801	-0 1801	-0 5501	-0.4022	-0.4761	76.9	-0.0871	7	-0 3757	
50.0	-0 4922		54.0	-0 4022	-0.4022	-0 6981	-0.3282	-0 4761	83.4	-0.1260	8	-0.3598	
56 0	-0 6381	-0 5286	65.0	-0 6715	-1 0222	-0 8079	-0 6715	-0 6520	89.4	-0.1845	9	-0.3439	
65 0	-1 0212	-0 6199	78.5	0.0103	-2.4638	-1 0416	-1 1391	-1.1780	95.4	-0.3014	10	-0 3598	
76 0	-0.3462	10.8923	79.5	-25.6157	-24.0750	-0.9982	-1 0509	-1.1695	101.1	-0.2624	11	-0.3757	
79.0	-3.7943	-0.5651	80.5	-24.0750	-1.9598	-0.9719	-1.0773	-0.3660	---BOTTOM---			12	-0.3439
80.5	-27.2494	2.4001	81 3	-26.6567	117.7275	-0.9851	-1.0773	-1.2485	38.2	0.0883			
81 0	-26 1957	0.7404	82.0	-23.8772	-3.3296	-1.0245	-1.0509	-1.2221	43.4	0 1272			
82 0	-25 6685	-1 6700	84 0	-21.9408	-1 7885	-0.8929	-1.0509	-1.2353	50.4	-0.1066			
84 0	-14 1040		87.0	-6 8992	-1 4461	-1 0245	-1.0641	-0.3528	56.1	-0.5352			
87 0	-18 3715		89.0	0 6233	-1.5015	-1 0892	-1.1368	-1.2954	62.5	-0.9442			
89 0	-5 6002	-2 0741	93.0	0 8770	-1.1843	-1.1527	-0.3915	-0.3757	69.7	-0.6130			
93 0	3.1857		96.0	-0.3123	-1.1051	-1 0099	-1.2002	-1.2478	76.9	-0.2040			
96 0	2 1727	0.5948	100.0	-0.7563	-1 1209	-0 8038	-1.0575	-0.3281	83 4	0.0688			
100 0	0 3220	-1.3339	96 0	0 3062	0 3696	0 3537	0 3379	0 1952	89 4	0 1272			
96 0	0 4974	0 2051	84 0	0 4489	0 5599	-0.3439	0 6392	0 5440	95.4	-0 0286			
84 0	0.3220	-1.3339	73 5	0 3379	-0 3282	0.5598	0.5598	0.4859	101.1	-0.5352			
73 0	0 4974	0.2051	54 0	0 2639	0 1898	0.1898	0 1898	0 1158					
50 0	0.7896	0.5363	33.0	-0.4022	-0 3282	0.0419	0.0419	0.0419					
35 0	0 6207	0 5295	24.0	-0.4805	-0 4293	-0.0454	0.0825	0.0570					
25.0	0.1829	-0 9118	10.0	-0.4805	-0.6852	-0.1733	0.0825	0 1158					
10 0	0.1464	0.0917	5.0	-0.5316	-0 6084	-0 1989	0 1081	0.1158					
5 0	-0.0178	0.0552	2.5	-0.5316	-0.5573	-0 1989	0.2105	0.1898					
2 5	0.0187	0 2194											

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TABLE A48 (Continued)

RUN SEQ	296 4	CLAERD =	0.7493	CDAERD =	0 5916	DCLC =	0.2851	DWLC =	0.4074	BASEPR =	8.1905	PAGE	733
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
4 02	2142.03	2128.69	13.34	106.12	-0 05	2.66	0.333	0 453	0 786	87 0501	0.672180E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
0 05	1 44	0 22	-0 30	0 00	-0 01	-0.00	0.98	-0.20	0.94	0.53			

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **	
%X/C	BP=3.375 CP	BP=10 125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XLOC	CP	NO.	CP
0 0	-0 5114	-2.3512	0 0	-0 6286	-0.4995	0 0682	-1.3254	-3 2353	38.2	-0.3386	1	-0 5698
2 5	-0.9713	-1.7257	2.5	-0 3447	0 3780	-0.6028	-1.1447	-1 6867	43.4	-0.3386	2	0 1018
5.0	-0 8057	-1 2472	5 0	0.4812	0.2231	-0.4995	-0.8608	-1.1963	50.4	-0.1814	3	-0 5698
10 0	-0 1066	-1 0081	10 0	-0.3447	-0.0607	-0 4995	-0.7318	-0 9383	56.1	-0.0635	4	0 1018
15 0	-0 6953.	-0.8977	15 0	0 0682	-0 1640	-0.4995	-0 6286	-0 8092	62.5	-0 0242	5	-0 5698
25 0	-0.5849	-0 7137	24.0	-0.1124	-0 2931	-0 5253	-0 4995	-0 6544	69.7	-0.0832	6	0 0272
35 0	-0 5481	-0.6217	33.0	-0.2713	-0 4206	-0.4952	-0.5698	-0 6444	76.9	-0.1225	7	-0 4418
50 0	-0.6033		54 0	-0 4952	-0.4952	-0.7190	-0.1966	-0.6444	83 4	-0.1421	8	-0.4098
56 0	-0 7689	-0 6585	65.0	-0 6725	-1.0655	-0.8100	-0 6922	-0 7315	89 4	-0.2011	9	-0 4258
65 0	-1 1001	-0 7873	78 5	0 3098	-2 6372	-1 1441	-1.1244	-1 3405	95.4	-0.2993	10	-0 4098
76.0	-0 3825	9.4608	79.5-25	7953	-28.5444	-0 9951	-1 0748	-1.3272	101.1	-0.2207	11	-0 4258
79.0	-3 9335	-0.6401	80 5-24	5594	-2 4961	-0.9287	-1 0615	-0 3841		---BOTTOM---	12	-0.4258
80 5	-27.7473	1 8209	81.3-24	4267	118.6727	-0.9951	-1.0881	-1.3537	38.2	0.1330		
81.0	-26 7776	0 7716	82.0-24.	1215	-3 3728	-0 9154	-1.0482	-1.2607	43.4	0.1919		
82.0	-26.2729	-1.8452	84 0-21	9962	-2 4430	-0 9420	-0 9819	-1.3139	50.4	-0.0242		
84 0	-15 1016		87.0	-8 9516	-1 7655	-0.9686	-1.1412	-0 3708	56.1	-0.4565		
87.0	-18 6217		89 0	-0 5697	-1 6731	-1 1454	-1.1614	-1.4013	62 5	-0 8690		
89 0	-4 7000	-1 9692	93 0	0 8055	-1 5452	-1.1454	-0 4418	-0 4258	69.7	-0.5547		
93 0	2.8244		96.0	-0 4258	-1 3533	-1 0655	-1.2254	-1.4013	76.9	-0.0635		
96.0	1 8814	0.7223	100.0	-0 8895	-1 2893	-0 9216	-1 1135	-0.4258	83.4	0.1526		
100 0	0 6241	-1 7924	96 0	0 2778	0 3577	0.3577	0.3257	0 1658	89 4	0.2312		
96 0	0 5652	0 1722	84 0	0 5336	0.7735	-0.4418	0 7095	0 5496	95.4	0.0151		
84 0	0 6241	-1 7924	73 5	0 3257	-0 3459	0 6242	0 6242	0 4750	101 1	-0.6137		
73 0	0 5652	0 1722	54.0	0 2511	0 2511	0 2511	0 2511	0 1018				
50 0	0 8206	0 6044	33 0	-0 1220	-0 3459	0 1018	0 1018	0.1018				
35 0	0 6294	0 5374	24 0	-0 5511	-0 3189	0 0166	0.1973	0 0940				
25.0	0 2614	-0 9897	10.0	-0 6544	-0.8608	0 0682	0 2747	0 2511				
10.0	0 2062	0 1510	5 0	-0 5511	-0 8867	0 1199	0 4038	0 3257				
5 0	0 0958	0 1327	2 5	-0 5511	-0 8608	0.1457	0 4812	0 4750				
2 5	0 3902	0 4638										

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TABLE 48. (Concluded)

RUN SEQ	296 6	CLAERO =	1 3521	CDAERO =	0.9399	DCLC =	0.3161	DWLC =	0.4451	BASEPR =	8.1905	PAGE	735
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
12.08	2141.96	2128 73	13 22	105 76	-0 05	2 74	0.343	0 468	0.811	89.5966	0.667901E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
0 05	2.11	0.66	-0.37	0.02	0 01	0.02	1.72	-0 26	1.57	0.90			

\*\*\*\*\* CANARD \*\*\*\*\*

%X/C	BP=3.375	BP=10 125
	CP	CP
0 0	-7 8076	-2.5196
2.5	-3.5957	-2.4824
5.0	-1.8516	-2 4824
10.0	-0 1446	-2 5567
15.0	-1.2208	-2.5937
25 0	-0.9424	-2 9463
35 0	-0.8867	-2 7607
50.0	-0.8125	
56.0	-0 8497	-0.9053
65 0	-1 0352	-0 5528
76 0	-0 6270	1.3955
79.0	-2.3711	-1.1651
80 5	-3 7363	1.7024
81.0	-3.3478	0 6843
82 0	-3.0263	-4 9553
84 0	-1 6868	
87 0	-3 0665	
89 0	-2 9765	-1 6689
93 0	-2 9170	
96.0	-3 2341	0 7483
100 0	0 7681	-0.4802
96 0	0 0945	0 3718
84.0	0 7681	-0 4802
73 0	0 0945	0 3718
50 0	0 7681	0 6492
35 0	0 7089	0 5791
25 0	0 3378	-1 0538
10 0	0.3935	0 3378
5 0	0 3378	0 3750
2 5	0 6162	0 5420

\*\*\*\*\* WING \*\*\*\*\*

%X/C	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
	CP	CP	CP	CP	CP
0.0	1 8907	0.6935	-2.8202	-5.0325	-2.2216
2 5	-0.5819	0 0688	-1 5188	-5.1626	-2.4818
5.0	0 3031	-0.2435	-1 2065	-3 8092	-2 4038
10.0	-0 5558	-0.5819	-1 1284	-1 1544	-2.5078
15.0	-0 5038	-0.6339	-1.0764	-1.0243	-2.5599
24.0	-0 4517	-0.7119	-1.0243	-0.8681	-2.5078
33.0	-0 4994	-0.6499	-0.9509	-0.8757	-2.2303
54.0	-0.6499	-0.8003	-1.0262	-0 1230	-1.1767
65 0	-0.9754	-1 1934	-1.2132	-1.0349	-1.0547
78 5	0.2926	-4.7002	-1 8472	-1 6689	-1 5897
79.5-26	6566	-39.9720	-1.7939	-1.6198	-1.7537
80 5-25	7324	-3 6023	-1 7805	-1.7403	-0.6552
81 3-26	0002	119.6249	-1.7135	-1.7135	-1.7135
82 0-25	7453	-5 2232	-1.7671	-1 6331	-1.6599
84 0-24	9550	-3 9105	-1.6198	-1.6465	-1.7269
87.0-16	3821	-2 9058	-1 8475	-1 7001	-0 6285
89 0 -6	9122	-2 8001	-1.9614	-1 7518	-1 7840
93 0 0.6833		-2.3646	-1 8808	-0 6713	-0.6390
96 0 -0	8326	-2.1388	-1 8163	-1 7840	-1 8002
100 0 -1	1390	-1.5260	-1 6389	-1.7195	-0.6068
96 0 1	1348	0 7478	0 3285	0.2479	0 0383
84 0 1.7798		1 5219	-0 6390	0 6994	0 3608
73.5 1	1563	-0 4994	0 7048	0.5543	0.3285
54 0 0	3285	0 3285	0 3285	0.3285	0.1779
33 0 0.	1027	-0 5746	0 3285	0.3285	0.1027
24 0 -0	0613	0 0688	0 3291	0.4332	0 2510
10.0 -0	3216	-0.0092	0 3812	0 5113	0 4037
5.0 -0.	6859	-0 2955	0 4072	0.5373	0.4790
2 5 -1.	2065	-0.7380	0.3812	0.3291	0.2532

\*\*FUSELAGE\*\*

XLOC	CP
38 2	-0.4802
43 4	-0 4405
50 4	-0.2820
56.1	-0.1829
62.5	-0.1631
69.7	-0.2226
76 9	-0 2622
83.4	-0.2424
89.4	-0.3415
95.4	-0.4207
101.1	-0.3415
---	BOTTOM---
38.2	0.1737
43.4	0.1935
50.4	-0.0442
56.1	0.0548
62.5	-0.0641
69.7	-0.0442
76.9	0.0548
83.4	0.1539
89 4	0 2529
95.4	-0.0442
101.1	-0.9556

\*\*MISC \*\*\*

NO.	CP
1	-0 8757
2	0 3285
3	-0 8757
4	0.3285
5	-2 2303
6	0 1027
7	-0.6390
8	-0.6390
9	-0.6229
10	-0.6552
11	-0.6390
12	-0.6068

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RUN SEQ	PROPLULSIVE WING / CANARD PRESSURES										PAGE 736			
297 1	CLAERD = 0 1632	CDAERD = 0 6370	DCLC = 0 6665	DWLC = 0 9592							BASEPR = 8.1905			
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN			
-0 05	2142 10	2136 56	5 54	68 42	-0 05	2 68	0 814	1 108	1.922	87.7528	0.431959E+06			
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR				
0.05	1.79	-0 39	-0 44	0 01	-0.01	-0 04	0.63	-0.18	0.64	0 45				
***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC **				
BP=3 375 BP=10 125			BP = 2 BP = 6 BP =12 BP =16 BP = 22					----TOP----						
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO	CP		
0 0	0 4614	-0 0258	0 0	-0 8089	-0 4983	-0 0012	0 3716	0 3095	38.2	-0.3314	1	-0.2647		
2 5	-0 4245	-0 9560	2 5	-0 2497	0 6823	-0 1876	-0 5604	-0 6847	43 4	-0.3787	2	-0.2647		
5 0	-0 4688	-0 7345	5 0	0 6823	0 6201	-0 1876	-0.3740	-0 5604	50.4	-0 2368	3	-0.4444		
10.0	-0 1587	-0 7345	10 0	-0.2497	0 4338	-0 2497	-0.2497	-0.4361	56.1	-0.0949	4	0.2743		
15 0	-0 4688	-0 7345	15.0	0 3095	0 1852	-0 3740	-0 1254	-0 4361	62 5	-0.0476	5	-0 4444		
25 0	-0 6017	-0 6460	24.0	0 1852	-0 0633	-0 4361	-0.1254	-0 4361	69 7	-0 0949	6	0 0946		
35 0	-0 6017	-0 6017	33 0	-0 0850	-0 0850	-0 4444	-0 4444	-0 4444	76.9	-0.1422	7	-0 4315		
50 0	-0 5574		54 0	-0 4444	-0 4444	-0 6240	-0 6240	-0 4444	83 4	-0.1422	8	-0 3545		
56 0	-0.7788	-0.6460	65.0	-0 8044	-1 6084	-0 8044	-0.7570	-0 7098	89 4	-0.2841	9	-0 3930		
65 0	-1 2219	-0 5574	78 5	-0 0949	-3.1691	-1 0881	-1 1354	-1 3246	95.4	-0.2841	10	-0.3930		
76 0	-0.3359	5.3337	79 5-62	2610	-62 7088	-0 8298	-1.0217	-1.2455	101.1	-0.2841	11	-0 3545		
79 0	-5 2967	-3 9680	80 5-58	3272	-2.8126	-0 8939	-0 9258	-0 1263	---BOTTOM---		12	-0 3545		
80 5	-53 6260	2 1122	81 3-58	1033	284 7031	-0 9578	-1 0217	-1.2775	38 2	0.0470				
81 0	-52 2829	0 8650	82 0-57	6231	-5 1149	-0 9258	-0.8619	-1 3416	43 4	-0 0003				
82 0	-51 7719	-2 5887	84 0-50	6851	-2.7806	-0 8298	-0 8619	-1 1497	50 4	-0.2841				
84.0	-26 1252		87.0	-6 6820	-2 0771	-0 8619	-0.8619	-0.0944	56.1	-1.3719				
87.0	-35 0147		89.0	4 5731	-2 0868	-1.1629	-1.2399	-1 4324	62.5	-1.6084				
89 0	-14 7570	-3 3110	93.0	1 6858	-1 5479	-1 2015	-0 4315	-0 3545	69 7	-0 9935				
93 0	0 2362		96 0	-0 4315	-1.3555	-1 0475	-1 2015	-1 6248	76.9	-0.3314				
96 0	1 1821	0 6145	100 0	-1.8559	-1 3555	-0 8164	-1 0859	-0 3930	83 4	-0 0476				
100 0	1 0875	-1.1354	96 0	0 1845	0 3385	0 3385	0 2999	0 1845	89 4	0.0470				
96.0	0 3780	0.1416	84 0	0 3385	0 5695	-0 3930	0 6849	0.5695	95.4	-0 1895				
84 0	1 0875	-1 1354	73.5	0 4540	-0.2647	0 6336	0.6336	0 6336	101 1	-0 8044				
73 0	0 3780	0 1416	54 0	0 2743	0 2743	0 2743	0 4540	0 2743						
50 0	0 7564	0 5200	33 0	-0 4444	-0 2647	0 0946	0 0946	0 0946						
35 0	0 5500	0 5057	24 0	-0 5604	-0 5604	-0 1876	0.0609	0 0609						
25 0	0 1513	-1 0888	10 0	-0 4983	-0 6225	-0 3740	0 1852	0 0946						
10.0	0 1513	0 0627	5 0	-0 6225	-0 5604	-0 3740	0 1852	0 2743						
5 0	-0 0702	0 0627	2 5	-0 5604	-0 6225	-0 4361	0 2473	0 2743						
2 5	-0 0258	0 2399												

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TABLE A49. (Continued)

RUN SEQ				PROPLULSIVE		WING / CANARD		PRESSURES			PAGE 738
297 3	CLAERO =	0.4234	CDAERO =	0.6985	DCLC =	0 6448	DWLC =	0 9196	BASEPR =	8.1905	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	HGT	RN	
3 99	2142.03	2135 92	6.10	71 84	-0 05	2.68	0.752	1.023	87.6917	0.453227E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	COTR	
0 05	1.99	-0.14	-0.41	0 01	-0 01	-0 04	0.92	-0.18	0.88	0.54	

\*\*\*\*\* CANARD \*\*\*\*\*

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.7692	-3 0204
2.5	-1 1713	-2.0154
5.0	-0.7692	-1.4526
10 0	-0.0859	-1.1310
15 0	-0.6890	-1.0104
25 0	-0.6486	-0.8095
35 0	-0 6085	-0 6890
50 0	-0.6085	
56.0	-0 8497	-0.6486
65 0	-1.2919	-0 5683
76.0	-0 3271	1 6426
79 0	-5.1508	-5.4323
80 5	-52.0226	2.2474
81.0	-51 2099	0.9414
82.0	-46.8567	-2.3090
84.0	-26.3095	
87.0	-35.2486	
89 0	-15.7402	-3.9368
93 0	0.9565	
96 0	3 2314	0 7848
100 0	1.5145	-1.3184
96.0	0.5273	0.3556
84.0	1.5145	-1.3184
73 0	0.5273	0 3556
50 0	0.8277	0 6560
35 0	0 6778	0 5171
25.0	0 1954	-1 0908
10 0	0 2357	0 1553
5 0	0 1151	0 1151
2 5	0.3161	0 5572

\*\*\*\*\* WING \*\*\*\*\*

	BP = 2	BP = 6	BP = 12	BP = 16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0 8050	-0.8050	-0 1283	-0.8613	-2.1019
2.5	-0 4102	0 2664	-0.5794	-1 1996	-2 2710
5.0	0.5483	0.2100	-0.5794	-0.8050	-1.2560
10.0	-0.4102	-0.0155	-0.6922	-0.6358	-0.9177
15 0	0 0972	-0.1283	-0.8050	-0.5230	-0.7486
24.0	-0.0719	-0.2975	-0.9742	-0.4666	-0 6922
33.0	-0.1419	-0.3050	-0 7941	-0.4680	-0.4680
54.0	-0 3050	-0.3050	-0.7941	-0.1419	-0.4680
65.0	-0 7174	-1.4472	-0 9321	-0.6316	-0 6745
78.5	0.6990	-3.4216	-1.1896	-1.1467	-1.1467
79.5	-56.2595	-61.2812	-1.0610	-1.1192	-1.2932
80 5	-52 8057	-6.3719	-1.3224	-0.9740	-0.3065
81.3	-52.8647	246.5798	-0.9740	-1.0900	-1 1482
82.0	-52.7779	-6.1398	-1 0900	-1.0320	-1.1772
84.0	-47.6399	-3 0926	-1.0320	-1.0900	-1.2062
87.0	-8.9839	-2.2800	-1 0030	-0.9740	-0 2195
89.0	3.7246	-1.7606	-1.2365	-1.2365	-1 3065
93.0	1.6632	-1 3762	-1.2016	-0.4330	-0.3282
96 0	-0 4680	-1 2016	-1 0618	-1.1318	-1.3762
100 0	-1 5860	-1.3065	-0.6426	-0.8872	-0.3632
96 0	0 2658	0 4404	0 4055	0 4055	0 1609
84 0	0.4404	0.6500	-0.3632	0.6850	0 5802
73.5	0 5103	-0.3050	0.6733	0 6733	0.5103
54.0	0 3472	0.3472	0 3472	0.3472	0.1842
33 0	-0 3050	-0 3050	0.1842	0.1842	0.1842
24.0	-0.7486	-0 5794	-0.1847	0 2664	-0.0155
10 0	-0.6358	-0.7486	-0 1847	0 2664	0 3472
5 0	-0.6358	-0 8050	-0.0719	0 3792	0.5103
2 5	-0 6358	-0 6922	0 0972	0 4356	0 5103

\*\*FUSELAGE\*\*

	-----TOP----
XLOC	CP
38.2	-0.3312
43 4	-0.3312
50.4	-0 1595
56.1	-0.0307
62.5	0.0122
69.7	-0.0307
76.9	-0.0736
83.4	-0 1166
89 4	-0.1595
95.4	-0.2024
101.1	-0.2024
	---BOTTOM---
38.2	0.1839
43.4	0.1839
50.4	-0.1166
56.1	-1.1896
62.5	-1.5330
69.7	-0.9321
76.9	-0.1595
83.4	0.1410
89 4	0 2268
95.4	-0 0736
101.1	-0.8462

\*\*MISC.\*\*

	NO.	CP
1	-0 4680	
2	0.0212	
3	-0.4680	
4	0.1842	
5	-0.4680	
6	0.0212	
7	-0.3981	
8	-0.3632	
9	-0.3981	
10	-0.3981	
11	-0.3981	
12	-0 3981	

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TABLE A49 (Concluded)

RUN SEQ 297 5  
 ALPHA 12 02  
 BETA 0 05

CLAERO = 0.9190  
 PTOT 2141 89  
 CL 2 70

PROPLULSIVE CDAERO = 0 9768  
 Q VEL 69 84  
 CM CROLL -0 32  
 -0.00

WING / CANARD DCLC = 0 7383  
 H/C 2.73  
 CN 0 02  
 -0.02

PRESSURES DWLC = 1 0383  
 CMUW 1 092  
 CMUT 1.894  
 CNTR 1 58  
 CMTR -0.07

PAGE 740  
 BASEPR = 8.1905  
 HGT RN 89.2687 0 439947E+06  
 CLTR CDR 1 43 0.88

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**	
%X/C	BP=3.375	BP=10.125	%X/C	BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----	NO	CP
	CP	CP		CP	CP	CP	CP	CP	XLOC		
0 0	-7.2933	-2.8670	0 0	-0.6656	-0.4865	-1.3819	-4.3074	-2.0984	38 2	-0.5845	1 -1.0655
2 5	-5.5484	-2.8670	2 5	-0.6059	-0.0089	-1.2029	-4.0685	-2.2179	43 4	-0.5391	2 0 3156
5 0	-1.7604	-2.9520	5 0	0.3494	-0.1880	-0.9641	-3.2924	-2.1581	50 4	-0.3573	3 -0 7202
10 0	-0.2282	-2.8670	10 0	-0.7253	-0.4865	-0.9044	-0.7850	-2.2179	56 1	-0.1755	4 0 3156
15 0	-1.4199	-3.0372	15 0	-0.0685	-0.4865	-0.9044	-0.7253	-2.3373	62.5	-0.1301	5 -1.9286
25 0	-1.2071	-3.2501	24 0	-0.3073	-0.6656	-0.9044	-0.6656	-2.2776	69 7	-0.1755	6 0 1429
35 0	-1.1646	-3.2074	33 0	-0.3750	-0.7202	-0.8929	-0.7202	-1.9286	76.9	-0.2209	7 -0.6216
50 0	-1.0794		54 0	-0.7202	-0.7202	-1.0655	-0.7202	-0.8929	83 4	-0.2209	8 -0.6585
56.0	-1.2922	-1.3348	65 0	-0.8572	-1.6753	-1.0390	-0.9026	-0.9481	89.4	-0.3119	9 -0.6216
65 0	-1.5901	-0.9942	78.5	1.0061	-3.9930	-1.4026	-1.4026	-1.4480	95 4	-0.4028	10 -0.6216
76 0	-0.6538	2.1126	79.5-59	9261	-73.8768	-1.3292	-1.2371	-1.4829	101.1	-0.3573	11 -0 6216
79 0	-5.8038	-3.2501	80 5-56	4237	-8.4888	-1.3599	-1.2371	-0.7146	---	BOTTOM---	12 -0.5845
80.5	-56.0855	1.8050	81.3-56	7308	245.2555	-1.4213	-1.2371	-1.5136	38 2	0.1880	
81.0	-54.4271	0.7603	82 0-56	3006	-7.9971	-1.2678	-1.2985	-1.5136	43.4	0.1880	
82 0	-53.9344	-4.8935	84 0-52	1216	-4.4326	-1.2062	-1.2371	-1.5443	50.4	-0.3119	
84 0	-30.5815		87 0-11	3771	-3.3880	-1.4213	-1.3906	-0.7455	56.1	-1.6298	
87 0	-40.1988		89 0 3	1516	-2.9149	-1.6203	-1.3984	-1.4354	62 5	-1.8116	
89 0	-13.3550	-2.1298	93 0 1	6719	-1.5464	-1.5464	-0.6585	-0.5845	69 7	-0.9481	
93 0	3.6420		96.0	-0.8804	-1.0284	-1.2874	-1.4354	-1.5464	76 9	-0.1301	
96 0	3.9147	0.7334	100.0	-1.7313	-0.7325	-1.0654	-1.3613	-0.6216	83 4	0.0517	
100 0	1.4605	-0.9481	96 0 0	3033	0.5622	0.4142	0.4142	0.2293	89 4	0.1426	
96 0	0.4607	0.1880	84 0 0	5992	0.8211	-0.6216	0.7471	0.5622	95.4	-0.2664	
84 0	1.4605	-0.9481	73 5 0	4882	-0.7202	0.6608	0.4882	0.4882	101 1	-1.4026	
73 0	0.4607	0.1880	54 0 0	3156	0.3156	0.3156	0.3156	0.1429			
50 0	0.7334	0.5516	33 0	-0.0297	-0.7202	0.3156	0.3156	0.3156			
35 0	0.6656	0.4528	24 0	-0.6656	-0.3073	0.1703	0.4091	0.1703			
25 0	0.2400	-1.5051	10 0	-0.6059	-0.8447	0.2896	0.5285	0.4882			
10 0	0.3677	0.2400	5 0	-0.5462	-0.7850	0.3494	0.6479	0.4882			
5 0	0.2826	0.2826	2.5	-0.5462	-0.7253	0.3494	0.5285	0.4882			
0 5	0.5379	0.4102									

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RUN SEQ 298 1 CL AERO = -0 2258 CDAERO = 0 8263 DCLC = 1 2037 DWLC = 1 7253 BASEPR = 8 1905 PAGE 741  
 ALPHA PTOT PSTAT Q VEL YAW H/C CMUC CMUW CMUT HGT RN  
 0 10 2142.10 2138.93 3.16 51 77 -0 05 2 69 1 468 1 990 3 458 87 8940 0.325344E+06  
 BETA CL CD CM CROLL CN CY CNTR CMTR CLTR CDTR  
 0 05 2 70 -1 01 -0 66 0 02 -0 01 -0 09 0 65 -0 21 0.65 0.46

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**			
%X/C	BP=3 375 CP	BP=10.125 CP	%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP	XLLOC	CP	NO.	CP	
0 0	0 3135	-0 2293	0.0	-0 7289	-0 5114	0.5763	0.5763	-0.1851	38.2	-0.1478	1	-0.2467	
2 5	-0.3068	-1.0822	2 5	-0.2939	0.9026	-0.1851	-0.2939	-0.9464	43.4	-0.2306	2	0.0679	
5 0	-0 4619	-1 0046	5 0	0 6850	0.5763	-0.1851	-0.0763	-0.7289	50.4	-0.0650	3	-0 5612	
10 0	-0.2293	-0 9271	10 0	-0 1851	0 3588	-0.2939	-0.1851	-0.6202	56.1	0 1005	4	0 0679	
15 0	-0 6170	-0 8496	15 0	0.3588	0 1412	-0.2939	0 0325	-0 5114	62.5	0.1005	5	-0.2467	
25 0	-0 5395	-0.6945	24 0	0 1412	-0 0763	-0.1851	0 0325	-0.2939	69.7	0.0178	6	0.0679	
35 0	-0 6170	-0 6170	33 0	0.0679	0 0679	-0.2467	-0 5612	-0 2467	76.9	0.1005	7	-0.4039	
50 0	-0 6170		54 0	-0 2467	-0 2467	-0.5612	-0 2467	-0 2467	83.4	-0.1478	8	-0.2691	
56 0	-0.9271	-0 6945	65 0	-0.7274	-2.0521	-0 8102	-0 6446	-0 7274	89.4	-0 1478	9	-0.3365	
65 0	-1 3924	-0 6170	78 5	0.5146	-4 1222	-1.1414	-0.9758	-1.4726	95.4	-0.2306	10	-0 3365	
76 0	-0.3843	3 3376	79.5	*****	-102 2755	-0.8920	-0 9480	-2.1794	101.1	-0.1478	11	-0.3365	
79 0	-6.9754	-4.0289	80.5	*****	-6.4903	-1.0039	-0.9480	-0 2762	---BOTTOM---			12	-0.3365
80 5	-110 6728	3.3626	81 3	*****	467.8423	-0 8920	-0 7241	-1.7877	38.2	0.1834			
81 0	-107 9295	1.0114	82.0	*****	-9 4573	-1.1719	-0.7241	-1.7877	43.4	0.1005			
82 0	-106 1944	-1.7877	84.0	-89 5673	-4 1949	-0.9480	-0.7800	-1.2278	50 4	-0.3134			
84 0	-57.2106		87 0	-7.5538	-2 6835	-0.8920	-0.7241	-0 1083	56.1	-2.5489			
87 0	-59.6156		89.0	9.7051	-2.0213	-1.1452	-1 2800	-1.6170	62.5	-1.9694			
89 0	-1 3898	-3 3769	93 0	3.3027	-1 6170	-1.0778	-0.4039	-0 2691	69.7	-0.8930			
93 0	15.1699		96 0	-0 4713	-1.4148	-1.0104	-1.2800	-1 7518	76.9	-0 2306			
96 0	8 9599	0.7630	100 0	-3.1671	-1 6170	-0.8082	-1.0778	-0.2691	83.4	0.1005			
100 0	3 2470	-1.3898	96 0	0 2026	0 4722	0 3374	0.2701	0 3374	89.4	0.2662			
96 0	0 4318	0 1834	84 0	0 4722	0 6744	-0 3365	0.6744	0 6070	95.4	-0.1478			
84 0	3 2470	-1 3898	73 5	0 3823	-0 2467	0.6968	0 6968	0 6968	101 1	-0.7274			
73 0	0 4318	0 1834	54 0	0.0679	0 3823	0.3823	0 0679	0 0679					
50 0	0 9285	0 6802	33 0	-0 5612	0.0679	0 0679	0 0679	0 0679					
35 0	0 6237	0 4686	24 0	-0 7289	-0 8377	-0 0763	0 2500	0 0325					
25 0	0 0034	-1 2373	10 0	-0 5114	-1 0552	-0 4026	0.3588	0 0679					
10 0	0.0809	0 0034	5 0	-0 4026	-0 7289	-0 1851	0 2500	0 0679					
5 0	0 0034	-0 0742	2.5	-0.4026	-0.8377	-0.1851	0.2500	0.0679					
2.5	-0 0742	0.3135											

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TABLE A50 (Continued)

RUN SEQ	298 3	CLAERO = -0 0367	CDAERO = 0.8375	DCLC = 1 2544	DWLC = 1.7834	BASEPR = 8.1905	PAGE 743
ALPHA	3 97	PTOT 2142 03	PSTAT 2138 86	Q VEL 3 17 51 80	YAW H/C -0 05 2 67	HGT RN 87.4363 0.325082E+06	
BETA	0.05	CL 3 00	CD -0 79	CM -0 63	CROLL 0 02	CN 0 02	CY -0.11
						CNTR 0.89	CMTR -0 17
						CLTR 0 85	CDTR 0.54

***** CANARD *****			***** WING *****					**FUSELAGE**		**MISC.**		
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO	CP
0 0	-0 7259	-3 2066	0.0	-0 5429	-0 5429	0 0008	-0.4341	-1.1953	38.2	-0.3449	1	-0.5926
2 5	-1 2686	-2 1213	2 5	-0 1079	0 5445	-0.3254	-1 0866	-1.5215	43.4	-0.3449	2	0.0363
5 0	-0 7259	-1 5786	5 0	0.7620	0.4358	-0.1079	-0.2167	-0 8691	50.4	-0.0138	3	-0.5926
10 0	0.0493	-1 2686	10 0	-0 1079	0 0008	-0.2167	-0 1079	-0 5429	56.1	0.1517	4	0.0363
15 0	-0 6484	-1 2686	15 0	0 3271	-0 3254	-0 2167	0.0008	-0 4341	62.5	0.0690	5	-0.5926
25 0	-0.5710	-0 9584	24 0	0.2183	-0 5429	-0.3254	0 0008	-0 3254	69.7	0.1517	6	0 0363
35 0	-0 5710	-0.6484	33 0	0 0363	-0 2782	-0.5926	-0 9071	-0 5926	76.9	-0.0138	7	-0.3006
50 0	-0.4158		54 0	-0 2782	-0 5926	-0 9071	0.3506	-0.5926	83 4	-0.0138	8	-0.2333
56 0	-0 9584	-0.5710	65 0	-0.6760	-1 9176	-0 9243	-0.6760	-0.6760	89 4	-0 0966	9	-0 1659
65 0	-1 2686	-0.4934	78 5	1.3106	-3.9870	-1 0899	-1 0071	-1.1726	95.4	-0.2621	10	-0.2333
76 0	-0 1834	-0 5710	79 5*****	-109 5545	-0 4197	-0.8114	-1.2032	101.1	-0.1793	11	-0.3006	
79 0	-6.8499	-6 3075	80 5*****	-8.7586	-0 5876	-0.4757	0.0840	---	BOTTOM---	12	-0 3006	
80 5	-109.0489	4 7852	81 3*****	443.1594	-0.5316	-0.6995	-0.7555	38 2	0.2345			
81 0	-105 4123	1.5391	82 0*****	-9.9899	-0.5316	-0.5876	-1.0913	43.4	0.2345			
82 0	-104 4044	-1 4830	84 0-90	0758	-4 0575	-0 4757	-0 6436	-0 9793	50.4	-0.3449		
84 0	-56.4424		87 0	-7.6951	-2 6025	-0.8674	-0.9234	-0 0279	56.1	-2.4143		
87 0	-60.3033		89 0	9 9403	-2.0523	-1 0417	-1.1092	-1.3787	62.5	-2 0832		
89 0	-5 1460	-3.8215	93 0	3 3376	-1 4460	-1.1092	-0.3006	-0 1659	69.7	-0.9243		
93 0	14.5545		96 0	-0 4354	-1 2439	-1 1766	-0.9744	-1.4460	76.9	-0 0966		
96 0	9 2569	0 6483	100 0	-3.0629	-1 5808	-0 8396	-0 9070	-0 3006	83 4	0.3173		
100 0	3 9594	-1 5866	96 0	0.3730	0 5078	0 3057	0.3730	0 3057	89 4	0.4000		
96 0	0 2345	0 2345	84 0	0 5752	0 8447	-0.5028	0.7773	0 7099	95 4	-0.1793		
84 0	3 9594	-1 5866	73 5	0 3506	-0 2782	0 3506	0.6650	0 6650	101 1	-0 9243		
73 0	0.2345	0 2345	54 0	0.0363	0 0363	0 0363	0 3506	0 0363				
50 0	0 6483	0 8140	33 0	-0 5926	-0 2782	-0.2782	0 0363	0 0363				
35 0	0.8244	0 6694	24 0	-0.6517	-0 9778	-0 2183	0.5445	0 2183				
25 0	0.2043	-1 1910	10 0	-0 4341	-1 3040	0.1095	0 6532	0 0363				
10 0	0 3593	0 2818	5 0	-0 2167	-0.7603	0 2183	0 6532	0.3506				
5 0	0.2818	-0 1058	2 5	-0 2167	-0.5429	0 2183	0.7620	0 6650				
2 5	0 5919	0 6694										

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TABLE A50. (Concluded)

RUN SEQ 298 5  
 ALPHA 12.04  
 BETA 0 05  
 CLAERO = 0.4507  
 PTOT 2141.96  
 PSTAT 2138.68  
 CD 3 59  
 CDAERO = 1 0036  
 Q 3.28  
 VEL 52 74  
 CM 0 02  
 DCLC = 1 3027  
 YAW -0.05  
 H/C 2.73  
 CN 0 01  
 DWLC = 1.8319  
 CMUC 1.415  
 CMUW 1.926  
 CMUT 3.341  
 CNTR 1.49  
 CMTR -0.10  
 BASEPR = 8 1905  
 HGT 89.4116  
 CLTR 0.330638E+06  
 RN  
 CDTR 0.83

***** CANARD *****			***** WING *****						**FUSELAGE**		**MISC **			
BP=3.375 BP=10.125			BP = 2		BP = 6		BP =12		BP =16		BP = 22			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	CP	CP	XLOC	CP	NO	CP
0 0	-6.2510	-2 5100	0.0	-0 5680	-0.2531	-2 6671	-4 0313	-1.9322	38.2	-0.3518	1	-0.8145		
2 5	-5 6525	-2 3606	2.5	-0 5680	0.1667	-1.5125	-3.9261	-2 0372	43.4	-0.3518	2	0.0960		
5 0	-2.5850	-2.4353	5 0	0.3766	-0.1482	-1 3026	-4.2411	-1.7224	50.4	-0.1121	3	-1.4215		
10.0	-0.1908	-2.2856	10.0	-0 5680	-0.4630	-1 1976	-0.5680	-1 8274	56 1	0.0476	4	0.0960		
15.0	-1.3878	-2.2856	15.0	-0.1482	-0.4630	-0.7778	-0.3580	-1.7224	62.5	0.0476	5	-1 4215		
25 0	-1 2382	-2.3606	24.0	-0.2531	-0.7778	-0 5680	-0.3580	-2.0372	69.7	0.0476	6	0.0960		
35.0	-1.1634	-2.1362	33 0	-0.5110	-0 8145	-0.8145	-1.1180	-2.0283	76.9	-0.0322	7	-0.4243		
50.0	-1 0886		54.0	-0 5110	-1.1180	-1.1180	-0 2076	-1 7248	83.4	0.0476	8	-0.3592		
56 0	-1.3131	-1 8367	65 0	-0.7513	-2.3490	-1.0708	-0.9110	-1.7898	89.4	-0.0322	9	-0.4243		
65 0	-1 6872	-1.6123	78.5	3.0834	-4.4263	-1 3903	-1.3903	-2.6685	95.4	-0.1121	10	-0.4893		
76 0	-0.4901	2.2033	79 5*****	-111 2920	-1.2083	-0.8842	-1.5324	101.1	-0.1121	11	-0.3592			
79 0	-7.2982	-8 7946	80.5-98	2740	-9.6886	-1.2624	-1.1003	-0 1820	---	---	12	-0.4243		
80 5	-110 9657	3 9771	81.3-98	.4905	410.2705	-1 2083	-0.8302	-2 1267	38.2	0.2074				
81 0	-107 4010	1.2224	82.0-98	.1642	-9.8509	-1 2083	-1 0463	-1.8565	43.4	0.2074				
82.0	-105.7268	-1.9105	84 0-88	6043	-5.0974	-0 8842	-0.8842	-1.6405	50 4	-0.4317				
84 0	-57.3293		87 0-10	6068	-3 9091	-0 9382	-1.0463	-0.2361	56.1	-2.3490				
87 0	-57 0610		89.0	9 3297	-3 4153	-1.3996	-1.5297	-1.7248	62.5	-2.2691				
89 0	1 4856	-2.0295	93 0	2 7620	-1 9849	-1 4647	-0.6193	-0 4243	69.7	-0.8311				
93 0	15.1468		96.0	-0 9445	-2.0499	-1.3996	-1.5297	-1 3996	76.9	0.2074				
96 0	8 2761	0 7667	100.0	-3 3504	-2.1801	-1 1396	-1.3996	-0.3592	83.4	0.3672				
100 0	3 5628	-1.7099	96.0	0.0309	0.4210	0 3560	0.2910	0 2910	89.4	0.2873				
96.0	0.3672	0 0476	84.0	0.5511	0.7461	-0 6193	0 8112	0 7461	95.4	-0 3518				
84 0	3 5628	-1 7099	73 5	0.3993	-0 5110	0.7028	0 7028	0 7028	101.1	-1.3903				
73 0	0.3672	0 0476	54.0	0 3993	0 3993	0.3993	0.0960	0.3993						
50 0	0.8466	0 6867	33.0	0.0960	-0 5110	0.0960	0 3993	0.3993						
35 0	0 7818	0 6322	24 0	-0 0432	0.0618	0 4815	0 9014	0 3766						
25 0	0.2581	-1 4627	10.0	-0 8827	-0 3580	0.5865	1 0063	0.3993						
10 0	0 4826	0.4078	5.0	-0 4630	-0.5680	0.9014	1.1112	0.7028						
5 0	0.1833	0 1833	2 5	-0.4630	-0 6729	1.0063	1.1112	0.7028						
2 5	0.5574	0 3329												

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#### A4.0 TABULATED DOWNWASH DATA

Table A51 presents the tabulation of downwash measurements behind the wing of the propulsive wing/canard. Downwash data are presented for the full span and 1/2 span blowing nozzles. The downwash of the short (1/4) span was not measured.

TABLE A51. TABULATED DOWNWASH DATA

RUN	POINT	$V_{\infty}$	VPROBE	$\epsilon$	CONFIG*	$\delta_F$	VARIABLE	$C_{\mu W}$	$C_L$	$C_{L_{TR}}$
57	11	157.16	97.52	1.92	BW6V	0	$\alpha=0$	0	0.142	
57	12	157.86	99.79	1.95			2	0	0.232	
57	13	158.52	100.07	2.95			4	0	0.322	
57	15	158.95	96.66	4.67			8	0	0.532	
57	17	159.05	98.66	9.52			12	0	0.752	
59	2	164.34	102.56	4.04	BW6V	0	$\alpha=0$	0.5	0.369	0.265
59	3	163.83	103.37	5.14			2	0.5	0.495	0.377
59	6	163.33	103.37	5.89			4	0.5	0.617	0.483
59	8	163.68	106.57	9.08			8	0.5	0.862	0.704
59	10	164.63	107.09	14.76			12	0.5	1.143	0.962
61	4	101.174	66.06	5.50	BW6V	0	$\alpha=0$	1.0	0.580	0.296
61	5	100.682	69.36	6.22			2	1.0	0.732	0.412
61	6	101.144	68.14	7.45			4	1.0	0.887	0.533
61	9	101.103	68.55	11.32			8	1.0	1.144	0.723
61	12	100.587	74.05	16.35			12	1.0	1.501	0.100
62	2	70.399	46.98	8.89	BW6V	0	$\alpha=0$	2.0	0.898	0.320
62	3	70.389	49.51	8.64			2	2.0	1.075	0.427
62	4	70.377	48.73	10.44			4	2.0	1.266	0.551
62	6	70.358	49.61	12.94			8	2.0	1.650	0.800
62	8	71.666	53.94	18.28			12	2.0	2.029	1.098
64	3	159.158	111.19	4.11	BW6V	15	$\alpha=0$	0	0.470	
64	4	159.230	111.44	5.20			2	0	0.560	
64	5	159.581	112.44	6.59			4	0	0.653	
64	7	157.348	114.17	9.60			8	0	0.845	
64	9	159.087	117.55	15.60			12	0	1.083	

\*CONFIG. code on Table 3

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu W}$	$C_L$	$C_{L\text{TR}}$
65	4	152.700	114.11	7.98	BW6V	15	$\alpha=0$	0.5	0.923	0.687
65	5	152.700	115.72	9.93			2	0.5	0.988	0.740
65	6	153.322	116.33	11.45			4	0.5	1.053	0.792
65	8	153.308	119.76	15.62			8	0.5	1.186	0.902
65	10	152.031	123.60	21.09			12	0.5	1.355	1.042
66	2	105.836	83.27	10.88	BW6V	15	$\alpha=2$	1.0	1.433	0.910
66	3	105.357	81.80	8.82			0	1.0	1.308	0.806
66	5	105.756	83.32	14.52			4	1.0	1.584	1.034
66	7	105.716	85.93	17.00			8	1.0	1.892	1.294
66	10	104.784	87.75	22.70			12	1.0	2.279	1.622
67	2	76.092	63.51	9.91	BW6V	15	$\alpha=0$	2.0	1.860	0.901
67	3	76.076	62.95	11.48			2	2.0	2.041	1.029
67	4	76.675	63.04	13.25			4	2.0	2.179	1.136
67	6	76.655	63.77	17.01			8	2.0	2.255	1.417
67	8	76.022	66.61	21.80			12	2.0	3.018	1.768
69	3	157.198	110.18	2.57	BC1W6V	15	$\alpha=0$	0	0.489	
69	4	157.884	109.93	1.25			2	0	0.592	
69	6	158.280	115.66	- 0.31			4	0	0.700	
69	8	158.358	143.29	27.10			8	0	0.938	
69	10	157.539	139.48	30.16			12	0	1.221	
70	2	151.288	115.73	6.20	BC1W6V	15	$\alpha=0$	0.5	1.099	0.792
70	3	151.608	114.91	6.90			2	0.5	1.246	0.922
70	4	151.906	125.83	8.31			4	0.5	1.381	1.042
70	6	156.534	134.22	8.59			8	0.5	1.656	
70	9	155.363	136.49	14.52			12	0.5	2.038	1.649

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu_w}$	$C_L$	$C_{L_{TR}}$
71	2	110.406	85.05	8.12	BC1W6V	15	$\alpha=0$	1.0	1.420	0.833
71	3	110.384	87.34	10.07			2	1.0	1.594	0.972
71	5	110.775	85.71	11.48			4	1.0	1.769	1.116
71	7	109.895	102.83	9.20			8	1.0	2.183	1.455
71	9	109.027	114.91	14.29			12	1.0	2.638	1.838
72	3	78.716	62.31	9.61	BC1W6V	15	$\alpha=0$	2.0	2.062	0.883
72	5	78.669	61.64	11.72			2	2.0	2.229	1.045
72	6	78.665	62.22	11.32			4	2.0	2.471	1.221
72	9	78.623	70.24	12.99			8	2.0	1.577	2.948
72	12	77.443	76.28	13.09			12	2.0	2.019	3.547
75	10	149.540	113.19	4.23	BC1W6V	45	$\alpha=0$	0	2.371	
75	11	148.960	117.31	2.33			2	0	2.535	
75	12	148.311	112.20	2.47			4	0	2.688	
76	2	148.624	116.45	8.06	BC1W6V	45	$\alpha=0$	0.5	1.578	1.049
76	3	150.850	123.54	9.55			2	0.5	1.641	1.118
76	4	150.576	123.08	12.15			4	0.5	1.756	1.221
76	7	149.156	132.45	15.70			8	0.5	2.079	1.523
76	10	149.540	141.47	18.59			12	0.5	2.371	1.805
77	2	105.958	85.68	8.35	BC1W6V	45	$\alpha=0$	1.0	2.330	1.273
77	3	105.509	84.81	10.09			2	1.0	2.469	1.386
77	4	105.962	85.04	13.51			4	1.0	2.585	1.495
77	6	104.656	92.92	17.84			8	1.0	2.960	1.822
77	8	102.856	100.85	19.22			12	1.0	3.459	2.256
79	3	74.745	59.82	9.13	BC1W6V	45	$\alpha=0$	1.0	3.456	1.298
79	9	74.750	60.62	14.47			8	1.0	4.110	1.870
79	11	74.139	71.08	18.58			12	1.0	4.614	2.304

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu_w}$	$C_L$	$C_{L_{\text{TR}}}$
85	2	158.756	114.66	5.13	BC2W6V	45	$\alpha=0$	0	0.514	
85	4	158.864	113.93	6.36			2	0	0.601	
85	7	159.301	118.97	10.30			8	0	0.917	
85	9	159.381	117.55	14.50			12	0	1.139	
86	1	150.652	120.53	7.48	BC2W6V	45	$\alpha=0$	0.5	1.543	1.037
86	2	150.023	119.40	8.39			2	0.5	1.615	1.084
86	3	151.610	124.14	11.63			4	0.5	1.675	1.147
86	4	153.514	121.23	10.24			4	0.5	1.657	1.149
86	5	152.911	125.90	15.44			8	0.5	1.945	1.436
86	6	150.731	143.52	20.68			12	0.5	2.222	1.727
86	7	151.082	144.70	21.91			12	0.5	2.218	1.677
87	1	104.471	87.42	8.74	BC2W6V	45	$\alpha=0$	1.0	2.649	1.608
87	2	104.412	84.46	9.64			2	1.0	2.761	1.702
87	4	103.928	84.77	10.80			4	1.0	2.857	1.774
87	5	103.467	85.80	12.51			8	1.0	2.982	1.863
87	6	101.585	95.25	18.98			12	1.0	3.367	2.185
88	1	74.305	62.35	6.65	BC2W6V	45	$\alpha=0$	2.0	3.950	1.918
88	2	72.376	59.71	8.43			2	2.0	4.217	2.029
88	4	74.898	60.09	9.39			4	2.0	4.140	2.037
88	5	74.229	58.58	13.68			8	2.0	4.301	2.106
88	7	73.583	63.99	17.54			12	2.0	4.617	2.337
89	2	149.841	122.26	11.31	BC2W6V	45	$h=14.3$	0.5	1.881	1.365
89	3	149.817	124.14	11.71			16.4	0.5	1.827	1.310
89	5	151.079	123.17	9.25			32.7	0.5	1.607	1.102
89	6	150.171	115.52	9.14			65.6	0.5	1.529	1.018
89	7	150.609	117.29	9.05			87.0	0.5	1.504	0.995

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu W}$	$C_L$	$C_{L_{TR}}$
90	1	75.349	60.62	11.70	BC2W6V	45	h=14.5	2.0	4.157	2.149
90	2	75.318	59.75	12.98			16.4	2.0	4.111	2.102
90	3	76.441	66.96	8.30			32.7	2.0	3.812	1.863
90	4	79.283	57.25	8.05			65.4	2.0	3.573	1.757
90	5	74.694	54.92	8.87			87.0	2.0	3.852	1.804
92	2	157.717	113.93	4.71	BC2W6V	45	h=14.2	0	0.741	
92	3	158.016	115.63	4.42			16.4	0	0.702	
92	4	156.860	108.91	6.25			32.7	0	0.602	
92	5	158.314	113.75	6.83			65.4	0	0.543	
92	6	158.591	111.70	6.52			87.0	0	0.524	
97	2	159.243	113.73	8.32	BC2W5V	45	$\alpha = 0$	0	0.577	
97	3	159.412	111.22	8.98			0	0	0.576	
97	4	159.822	111.86	10.71			2	0	0.663	
97	5	159.596	112.61	11.95			4	0	0.750	
97	6	159.982	115.65	14.55			8	0	0.891	
97	7	160.046	113.36	19.59			12	0	1.076	
98	1	143.023	124.91	8.24	BC2W5V	45	$\alpha = 0$	0.5	1.834	1.242
98	3	143.766	126.32	11.04			2	0.5	1.905	1.304
98	4	144.800	137.69	11.24			4	0.5	1.987	1.385
98	5	144.834	148.34	17.71			8	0.5	2.217	1.599
98	6	144.523	157.68	23.03			12	0.5	2.393	1.758
99	2	99.412	90.60	8.01	BC2W5V	45	$\alpha = 0$	1.0	3.088	1.855
99	4	101.778	92.31	10.50			2	1.0	3.111	1.914
99	6	99.834	93.63	11.33			4	1.0	3.295	2.034
99	7	100.766	104.14	13.56			8	1.0	3.485	2.212
99	8	98.838	108.76	18.55			12	1.0	3.761	2.413

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu W}$	$C_L$	$C_{L_{\text{TR}}}$
100	2	73.145	60.78	6.89	BC2W5V	45	$\alpha=0$	2.0	4.332	2.042
100	3	71.161	59.04	8.42			2	2.0	4.587	2.127
100	4	72.457	59.41	7.27			4	2.0	4.581	2.174
100	5	72.481	63.81	11.02			6	2.0	4.720	2.281
100	6	71.798	66.82	9.69			8	2.0	4.885	2.370
100	7	73.106	74.45	16.28			12	2.0	5.062	2.586
108	3	159.033	109.16	3.64	BC3W6V	45	$\alpha=0$	0	0.544	
108	4	159.396	109.93	3.59			2	0	0.647	
108	6	160.087	94.61	8.76			8	0	0.962	
108	9	159.846	110.55	28.23			12	0	1.199	
109	4	152.576	118.25	8.19	BC3W6V	45	$\alpha=0$	0.5	1.408	0.935
109	5	152.565	120.10	9.96			2	0.5	1.496	1.019
109	7	152.273	120.74	12.35			4	0.5	1.586	1.094
109	12	148.460	136.04	16.83			12	0.5	2.128	1.576
110	2	103.301	82.34	8.79	BC3W6V	45	$\alpha=0$	1.0	2.368	1.364
110	3	103.727	81.30	11.06			2	1.0	2.523	1.512
110	5	105.510	86.35	13.44			4	1.0	2.646	1.657
110	6	104.616	93.96	13.20			8	1.0	2.990	1.950
110	7	105.019	100.78	21.72			12	1.0	3.286	2.202
111	2	75.707	58.21	10.64	BC3W6V	45	$\alpha=0$	2.0	3.374	1.466
111	3	75.061	57.32	10.18			2	2.0	3.575	1.599
111	5	75.036	58.32	10.97			4	2.0	3.786	1.778
111	8	75.015	65.95	14.93			8	2.0	4.345	2.282
111	10	75.630	76.21	19.86			12	2.0	4.704	2.625

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu_w}$	$C_L$	$C_{L_{TR}}$
142	2	160.726	112.19	6.28	BW6V	45	$\alpha=0$	0	0.528	
142	3	161.723	113.92	6.50			2	0	0.615	
142	4	162.090	113.92	7.89			4	0	0.704	
142	5	161.542	112.19	12.00			8	0	0.883	
142	6	161.597	119.95	19.34			12	0	1.088	
143	1	129.940	121.41	8.88	BW6V	45	$\alpha=0$	0.5	1.462	1.080
143	2	129.570	122.13	11.28			2	0.5	1.490	1.100
143	3	131.750	124.91	11.69			4	0.5	1.498	1.116
143	4	131.710	131.11	16.01			8	0.5	1.570	1.179
143	5	133.756	136.07	20.08			12	0.5	1.629	1.242
144	1	88.261	88.95	9.64	BW6V	45	$\alpha=0$	1.0	2.564	1.751
144	2	87.153	87.79	13.27			2	1.0	2.763	1.915
144	3	88.759	91.27	12.53			4	1.0	2.851	2.021
144	4	87.540	91.12	17.89			8	1.0	3.243	2.372
144	5	87.569	92.75	20.71			12	1.0	3.405	2.512
145	1	65.763	61.99	12.76	BW6V	45	$\alpha=0$	2.0	3.439	1.991
145	2	64.344	63.04	10.53			2	2.0	3.652	2.105
145	3	63.569	63.77	12.30			4	2.0	3.834	2.225
145	4	65.081	66.08	15.28			8	2.0	4.122	2.538
145	5	64.329	67.74	19.23			12	2.0	4.423	2.777
146	1	63.788	53.70	9.76	Bw6V	45	$\alpha=0$	3.0	4.499	2.276
146	2	64.528	55.10	11.01			2	3.0	4.588	2.380
146	3	64.522	56.92	11.43			4	3.0	4.773	2.533
146	4	63.755	56.73	16.58			8	3.0	5.238	2.891
146	5	63.762	57.72	19.26			12	3.0	5.477	3.088

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu W}$	$C_L$	$C_{L_{\text{TR}}}$
147	1	54.752	45.49	9.02	BW6V	45	$\alpha = 0$	4.0	5.497	2.465
147	2	54.749	46.32	10.53			2	4.0	5.672	2.595
147	3	55.620	46.72	10.97			4	4.0	5.689	2.669
147	4	56.480	49.45	14.12			8	4.0	5.935	2.939
147	5	55.616	50.38	17.89			12	4.0	6.420	3.271
148	1	67.401	57.52	9.48	BW6V	45	$\alpha = 0$	2.0	3.496	2.083
148	2	67.391	58.29	12.74			2	2.0	3.663	2.227
148	3	67.383	54.24	14.57			4	2.0	3.800	2.343
148	4	68.104	58.47	16.12			8	2.0	4.197	2.659
148	5	67.376	61.29	18.26			12	2.0	4.534	2.932
149	1	67.389	54.24	10.68	BW6V	45	$\alpha = 0$	1.0	2.490	1.912
149	2	67.383	54.93	12.19			2	1.0	2.624	2.034
149	3	68.093	55.75	13.92			4	1.0	2.685	2.101
149	4	67.370	57.87	18.10			8	1.0	3.031	2.419
149	5	67.370	59.96	21.27			12	1.0	3.358	2.670
153	1	76.777	46.24	13.54	BW6V	45	$h=14.7$	2.0	2.154	0.638
153	2	76.154	50.32	8.79			16.8	2.0	2.513	0.973
153	3	76.095	58.95	12.36			32.7	2.0	3.530	1.989
153	4	76.729	58.21	7.73			65.5	2.0	3.465	1.946
153	5	76.123	58.99	7.50			87.7	2.0	3.468	1.923
154	1	155.226	115.49	9.92	BW6V	45	$h=15.3$	0.5	0.984	0.600
154	2	154.628	115.90	10.82			16.5	0.5	0.961	0.581
154	3	153.611	122.99	9.27			32.7	0.5	0.831	0.455
154	4	151.573	119.20	7.96			65.4	0.5	0.815	0.429
154	5	151.350	118.52	8.23			86.9	0.5	0.816	0.425

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{H_W}$	$C_L$	$C_{L_{TR}}$
155	1	160.556	110.94	4.49	BW6V	45	h=14.9	0	0.723	
155	2	160.883	110.44	4.86			16.5	0	0.701	
155	3	160.013	109.42	4.84			33.3	0	0.606	
155	4	160.313	108.91	6.11			65.7	0	0.552	
155	5	160.362	109.16	5.35			87.1	0	0.530	
156	1	161.623	110.69	5.37	BW6V	45	h=14.9	0	0.727	
156	2	161.651	111.45	4.94			16.5	0	0.706	
156	3	161.067	109.42	5.86			32.6	0	0.609	
156	4	161.779	113.61	7.22			65.5	0	0.550	
156	5	160.204	110.69	6.42			87.3	0	0.526	
157	1	154.846	113.29	11.23	BW6V	45	h=15.5	0.5	0.998	0.626
157	2	154.213	114.61	12.22			16.4	0.5	0.982	0.607
157	4	154.248	124.48	10.06			32.6	0.5	0.828	0.447
157	5	152.819	121.66	8.83			65.6	0.5	0.789	0.395
157	6	154.085	121.11	8.89			87.3	0.5	0.764	0.377
158	1	106.073	78.39	11.69			BW6V	45	h=15.6	1.0
158	2	106.456	79.66	12.92	16.4	1.0			2.052	1.248
158	3	107.147	89.87	10.34	32.6	1.0			1.883	1.088
158	4	107.154	85.91	8.45	65.6	1.0			1.814	1.018
158	6	106.308	85.54	10.89	87.0	1.0			1.810	0.999
159	1	78.740	56.33	8.93	BW6V	45			h=14.9	2.0
159	2	77.613	50.67	11.09			16.4	2.0	2.536	1.007
159	3	76.241	64.61	11.38			32.9	2.0	3.668	2.095
159	4	77.458	62.63	10.30			65.5	2.0	3.559	2.029
159	5	76.856	61.93	11.53			86.9	2.0	3.558	2.002

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TABLE A51. TABULATED DOWNWASH DATA

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu W}$	$C_L$	$C_{L_{TR}}$
160	1	76.226	64.45	16.38	BW6V	45	h=30.4	2.0	3.899	2.269
160	2	76.817	62.80	15.63			32.6	2.0	3.850	2.245
160	3	76.817	64.63	12.61			65.5	2.0	3.575	2.039
160	4	78.000	64.45	14.27			87.2	2.0	3.600	2.040
161	1	104.795	86.85	16.69	BW6V	45	h=30.4	1.0	2.140	1.269
161	2	107.436	90.46	15.88			32.4	1.0	2.024	1.192
161	3	108.308	88.48	12.67			65.5	1.0	1.894	1.078
161	4	107.033	87.48	13.92			87.1	1.0	1.900	1.065
162	1	151.796	124.64	16.64	BW6V	45	h=30.4	0.5	0.916	0.520
162	2	152.435	126.17	15.18			32.8	0.5	0.892	0.499
162	3	153.404	122.49	13.17			65.8	0.5	0.814	0.425
162	4	153.291	123.13	12.45			87.0	0.5	0.806	0.417
163	1	169.088	111.45	8.22	BW6V	45	h=29.4	0	0.808	
163	2	160.708	111.20	7.87			32.9	0	0.797	
163	3	160.120	110.44	8.24			65.3	0	0.729	
163	4	161.011	111.20	8.20			87.1	0	0.702	
								$C_{\mu C}$		
171	2	157.605	118.97	- 2.67	BC1V	45	$\alpha=0$	0	0.201	
171	3	157.701	116.84	- 2.54			2	0	0.239	
171	4	157.488	110.94	3.41			4	0	0.279	
171	5	157.872	105.00	23.62			8	0	0.348	
171	6	157.985	115.87	13.51			12	0	0.374	
172	1	150.558	108.39	- 2.14	BC1V	45	$\alpha=0$	0.25	0.487	
172	2	149.074	111.70	- 2.00			2	0.25	0.531	
172	3	149.454	119.68	- 3.09			4	0.25	0.562	
172	5	149.817	142.64	23.99			8	0.25	0.663	
172	6	149.540	139.61	17.36			12	0.25	0.837	

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu C}$	$C_L$	$C_{L_{TR}}$
173	1	102.061	72.13	- 0.59	BC1V	45	$\alpha=0$	0.5	0.724	0.435
173	2	102.521	73.29	- 1.58			2	0.5	0.769	0.491
173	3	102.890	75.54	- 1.37			4	0.5	0.827	0.558
173	4	102.964	92.69	- 1.06			8	0.5	0.926	0.674
173	5	102.506	81.93	4.54			12	0.5	1.025	0.788
174	1	74.926	52.36	1.99	BC1V	45	$\alpha=0$	1.0	0.971	0.430
174	2	70.393	50.73	- 0.42			2	1.0	1.083	0.486
174	3	70.370	50.17	- 0.31			4	1.0	1.141	0.560
174	4	72.956	53.42	1.85			8	1.0	1.182	0.692
176	2	158.081	112.20	- 2.23	BC1V	45	$\alpha=0$	0	1.115	1.092
176	3	158.123	107.61	- 2.45			2	0	1.466	1.444
176	4	158.496	107.61	8.77			4	0	1.778	1.755
176	5	158.571	111.70	14.37			8	0	0.252	0.249
176	6	158.355	114.66	9.13			12	0	0.343	0.341
177	1	143.431	106.04	- 2.24	BC1V	45	$\alpha=0$	0.25	0.427	0.303
177	2	146.725	110.69	- 3.44			2	0.25	0.465	0.344
177	3	146.081	123.86	- 3.53			4	0.25	0.519	0.394
177	4	148.665	147.57	- 0.85			8	0.25	0.578	0.452
177	5	148.378	137.84	30.28			12	0.25	0.754	0.622
178	1	102.817	73.29	- 0.62	BC1V	45	$\alpha=0$	0.5	0.596	0.355
178	2	102.788	73.67	- 1.15			2	0.5	0.640	0.392
178	3	102.308	73.29	- 0.78			2	0.5	0.642	0.391
178	4	102.746	77.01	- 0.38			4	0.5	0.694	0.440
178	5	103.651	99.16	2.11			8	0.5	0.810	0.548
178	6	106.320	77.01	11.82			12	0.5	0.938	0.679

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TABLE A51. TABULATED DOWNWASH DATA

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu_C}$	$C_L$	$C_{L_{\text{TR}}}$
179	1	74.097	50.73	1.17	BC1V	45	$\alpha=0$	1.0	0.819	0.352
179	2	72.814	48.47	2.49			2	1.0	0.898	0.401
179	3	69.533	50.73	1.58			4	1.0	1.014	0.456
179	4	70.187	56.96	3.18			8	1.0	1.114	0.542
179	5	70.181	71.35	5.59			12	1.0	1.280	0.687
181	3	157.599	111.70	- 0.15	BV	0	$\alpha=0$	0	-0.000	-0.038
181	4	158.035	115.63	2.39			2	0	-0.000	-0.033
181	5	157.780	118.97	- 0.46			4	0	0.104	-0.027
181	6	157.831	121.99	- 0.55			8	0	0.022	-0.015
181	7	157.583	125.85	- 2.04			12	0	0.039	0.002
								$C_{\mu_W}$		
187	2	156.237	128.68	6.98	BW6V	45	$\alpha=0$	0	0.546	
187	3	156.939	128.33	7.93			2	0	0.636	
187	4	157.007	127.43	9.66			4	0	0.720	
187	5	157.097	125.92	12.41			8	0	0.911	
187	6	156.262	124.39	20.65			12	0		
188	1	155.868	130.67	7.01	BW6V	45	$\alpha=0$	$b_N=1/2$ 0.5	0.885	0.627
188	2	153.483	129.06	8.02			2	0.5	0.964	0.692
188	3	154.131	130.34	9.23			4	0.5	1.013	0.738
188	4	158.090	133.78	14.00			8	0.5	1.080	0.810
188	5	155.132	149.57	21.90			12	0.5	1.200	0.912
189	1	107.023	99.72	7.41	BW6V	45	$\alpha=0$	$b_N=1/2$ 1.0	1.527	0.977
189	2	109.171	101.22	9.55			2	1.0	1.587	1.047
189	3	107.427	96.56	15.39			8	1.0	1.872	1.244
189	4	110.382	108.82	20.90			12	1.0	1.970	1.400

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu W}$	$C_L$	$C_{L\text{TR}}$
190	1	77.350	73.37	6.26	BW6V	45	$\alpha = 0$	$b_N = 1/2$ 2.0	2.142	1.088
190	2	77.746	72.82	9.14			2	2.0	2.279	1.186
190	3	77.927	73.85	8.95			4	2.0	2.330	1.250
190	4	77.337	77.30	15.95			8	2.0	2.627	1.495
190	5	76.732	77.76	17.96			12	2.0	2.800	1.619
216	2	128.442	104.65	7.59	BC1W6V	45	$\alpha = 0$	0	0.526	0.433
216	3	128.477	107.87	5.19			2	0	0.650	0.556
216	4	128.864	113.07	5.50			4	0	0.759	0.665
216	5	128.888	89.89	19.02			8	0	0.996	0.901
216	7	128.566	113.07	34.54			12	0	1.226	1.131
217	1	80.186	73.71	9.82	BC1W6V	45	$\alpha = 0$	$b_N = 1/2$ 1.0	2.023	0.932
217	3	80.167	73.72	8.94			2	1.0	2.003	0.892
217	4	80.155	73.08	9.27			4	1.0	2.076	0.943
217	5	79.563	79.67	11.81			8	1.0	2.434	1.246
217	6	78.405	93.10	14.33			12	1.0	2.893	1.632
218	1	57.454	39.95	5.14	BC1W6V	45	$\alpha = 0$	$b_N = 1/2$ 2.0	3.292	1.160
218	2	55.827	38.61	9.49			2	2.0	3.510	1.200
218	3	56.622	36.68	10.17			4	2.0	3.412	1.122
218	4	58.215	36.10	11.88			8	2.0	3.434	1.188
218	5	56.610	40.97	12.82			12	2.0	4.018	1.571

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TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu W}$	$C_L$	$C_{L_{TR}}$	
227	1	60.963	52.89	5.46	BW6V	45	h=14.3	$b_N=1/2$ 2.0	2.104	0.655	
227	2	61.674	56.64	7.13				16.4	2.0	2.770	1.321
227	3	65.085	48.35	9.85				32.6	2.0	2.668	1.369
227	4	60.253	43.83	5.70				65.6	2.0	2.765	1.250
227	5	59.489	40.03	10.87				86.7	2.0	2.817	1.266
228	1	83.702	68.75	9.39	BW6V	45	h=14.4	$b_N=1/2$ 1.0	1.913	1.176	
228	2	84.223	68.79	11.78				16.3	1.0	2.034	1.261
228	3	90.182	68.27	8.28				32.6	1.0	1.824	1.144
228	5	87.728	63.91	8.15				65.5	1.0	1.801	1.084
229	1	116.119	94.92	9.84	BW6V	45	h=14.8	$b_N=1/2$ 0.5	1.582	1.239	
229	2	111.035	84.76	9.21				16.3	0.5	1.659	1.281
229	3	114.876	89.20	6.46				32.7	0.5	1.478	1.126
229	5	112.610	87.10	6.46				64.7	0.5	1.428	1.062
229	6	115.743	87.65	7.00				87.2	0.5	1.385	1.039
230	1	128.458	93.75	8.65	BW6V	45	h=14.8	0	0.762	0.816	
230	2	128.446	94.15	8.69				16.3	0	0.742	0.796
230	3	128.089	94.56	7.23				32.7	0	0.641	0.696
230	4	126.983	95.67	7.80				65.3	0	0.582	0.638
230	5	126.668	94.74	6.73				86.9	0	0.565	0.622
231	1	127.604	92.52	14.25	BW6V	45	$\alpha=8$ h=43.7	0	1.006	1.063	
231	2	127.545	93.42	13.30			64.8	0	0.949	1.006	
231	3	127.024	95.50	12.64			86.8	0	0.928	0.986	

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TABLE A51. TABULATED DOWNWASH DATA (Concluded)

RUN	POINT	$V_{\infty}$	$V_{\text{PROBE}}$	$\epsilon$	CONFIG.	$\delta_F$	VARIABLE	$C_{\mu W}$	$C_L$	$C_{L\text{TR}}$
232	2	114.951	97.45	17.04	BW6V	45	$\alpha = 8$	0.5	1.785	1.405
232	4	114.968	91.63	14.18			$h=44.5$			
232	5	115.076	88.92	14.92			65.4			
							87.1	0.5	1.686	1.302
233	1	77.974	67.83	15.47	BW6V	45	$\alpha = 8$	1.0	2.618	1.791
233	2	78.993	63.06	14.52			$h=44.5$			
233	3	78.434	61.75	17.31			65.5			
233	4	57.364	51.41	15.53			87.0			
233	7	58.113	45.47	14.96			44.5			
233	9	58.091	45.13	16.78			65.2			
							86.7	2.0	3.298	1.833
								2.0	3.251	1.792

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## A5.0 TABULATED RUN INDEX

Table A52 presents the Run Index for NASA-LaRC Test 290.

TABLE A52. RUN INDEX

PREPARED BY  
 CHECKED BY  
 DATE 26 JAN 84

NASA-LaRC  
 TEST 290  
 14.5 x 21.75 VSTOL  
 FULL-SPAN  
 PROPULSIVE  
 WING

A-377

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	$S_c$	$S_{Fw}$	$S_{Fc}$	SCV DATA	DOWN WASH	H/C	NPR	$C_{Mw}$	$C_{Mc}$	Q	$\alpha$	$\beta$	REMARKS	TIME STARTED	TIME FINISHED	DATE
40	BW	1	1	6	-	-	0°	-			∞	A	∞	-	0	0°	0°	Noz Calib & Static Thrust	5:50		1/26
41	BC	-	1	-	1	0°	-	0°			∞	B	-	∞	0	0°	0°	" " " "	6:30		1/26
42	BWC	1	1	6	1	0°	0°	0°			∞	C	∞	∞	0	0°	0°	" " " "			1/27
43																		Alpha Calib Run ??			1/27
44	BWC	1	1	6	1	0°	0°	0°			∞	1.0	-	-	0		0°	Wt Tare			1/27
45	BWC	1	1	6	1	0°	0°	0°	Y		∞	C	∞	∞	0	0°	0°	Static Run	10:30		1/27
46	BWC	1	1	6	1	0°	0°	0°	Y		∞	1.0	0	0	30	A	0°		10:55		1/27
47	BWC	1	1	6	1	0°	0°	0°	Y		∞	1.0	0	0	30	A	0°				
48									Y									?			
49									Y									?			
50	BWC	1	1	6	1	0°	0°	0°	Y		∞	2.5	.5	.2	28	C	0°	Includes Wt Tare Run	5:00		1/30
51																		X-DUCER CALIB RUN			1/30
52	BWC	1	1	6	1	0°	0°	0°	Y		∞	2.5	.5	.2	28	D	0°	Cont'd from run 50			1/30
53	BWC	1	1	6	1	0°	0°	0°	Y		∞	2.5	1.0	.4	136	A	0°				1/30
54	BWC	1	1	6	1	0°	0°	0°	Y		∞	2.5	2.0	.8	71	A	0°				1/30
55	BWC	1	1	6	1	0°	0°	0°	Y		∞	1.0	0	0	30	A	0°				1/30
56	BW	1	-	6	-	-	0°	-	Y		∞	D	∞	-	0	0°	0°	STATIC RUN	9:50		1/31
57	BW	1	-	6	-	-	0°	-	Y		∞	1.0	0	0	30	A	0°	Includes Wt Tare	9:35		1/31
58	BW	1	-	6	-	-	0°	-			A	1.0	0	0	30	0°	0°	Grnd Ht Sweep	10:15		1/31
59	BW	1	-	6	-	-	0°	-	Y		∞	2.5	.5	0	32	A	0°		10:20		1/31

NOTES

	NPR	$\alpha$	H/C	Q
A	1, 6, 2, 4, 3, 1, 3, 6	-2° → 22° Δ=2°	4, .5, 1, 2, ∞	30, 24, 15, 8, 3
B	1, 6, 2, 3, 3, 0, 3, 6, 4, 3	0, 2, 4, 8, 12, 16, 20	6, 5, 1, 2, ∞	
C	1, 5, 2, 1, 2, 5	-2°, 0, 4°	9, 1, 2, ∞	
D	1, 5, 2, 2, 5, 3, 3, 5	4° → 22° Δ=2°		
E	1, 5, 2, 2, 5, 3	0 → 12°		

TABLE A52. (Continued)

PREPARED BY
CHECKED BY
DATE

NASA-LARC
TEST 290
14.5 x 21.75 VSTOL
FULL-SPAN PROPULSIVE WING

A-378

BLD #	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	S <sub>C</sub>	S <sub>F<sub>W</sub></sub>	S <sub>F<sub>C</sub></sub>	SCV DATA	DOWN WASH	H/C	NPR	C <sub>M W</sub>	C <sub>M C</sub>	Q	α	β	REMARKS	TIME STARTED	TIME PRODUCED	DATE
60									Y									RUN ABORTED - Tunnel failure			
61	BW	1	-	6	-	-	0°	-	Y		∞	2.5	1.0	0	12	A	0°		2.00		1/31
62	BW	1	-	6	-	-	0°	-	Y		∞	2.5	2.0	0	61	A	0°		2.45		1/31
63	BW	1	-	6	-	-	15°	-			∞	D	∞	-	0	0°	0°	STATIC RUN			1/31
64	BW	1	-	6	-	-	15°	-	Y		∞	1.0	0	-	30	A	0°				1/31
65	BW	1	-	6	-	-	15°	-	Y		∞	2.5	.5	0	28	A	0°				1/31
66	BW	1	-	6	-	-	15°	-	Y		∞	2.5	1.0	0	133	A	0°				1/31
67	BW	1	-	6	-	-	15°	-	Y		∞	2.5	2.0	0	7	A	0°				1/31
68	BWC	1	1	6	1	0°	15°	15°			∞	C	∞	∞	0	0°	0°	WT TARE & STATIC RUN			1/31
69	BWC	1	1	6	1	0°	15°	15°	Y		∞	1.0	0	0	30	A	0°				1/31
70	BWC	1	1	6	1	0°	15°	15°	Y		∞	2.5	.53	.22	29	A	0°				1/31
71	BWC	1	1	6	1	0°	15°	15°	Y		∞	2.5	1.0	.43	14.5	A	0°				1/31
72	BWC	1	1	6	1	0°	15°	15°	Y		∞	2.5	2.0	.85	7.5	A	0°				1/31
73	BWC	1	1	6	1	0°	45°	45°			∞	C	∞	∞	0	0°	0°	STATIC RUN	9.00		2/1
74	BWC	1	1	6	1	0°	45°	45°			∞	1.0	0	0	30		0°	RUN ABORTED - No WT TARE			2/1
75	BWC	1	1	6	1	0°	45°	45°	Y		∞	1.0	0	0	30	A	0°		10.10		2/1
76	BWC	1	1	6	1	0°	45°	45°	Y		∞	2.5	.5	.23	2.7	A	0°		10.45		2/1
77	BWC	1	1	6	1	0°	45°	45°	Y		∞	2.5	1.0	.5	13.3	A	0°		11.15		2/1
78	BVIC	1	1	6	1	0°	45°	45°			A	2.5	1.0	.5	13	0°	0°	Gnd Hgt Sweep			
79	BWC	1	1	6	1	0°	45°	45°	Y		∞	2.5	2.0	1.0	7	A	0°				

NOTES

A	
B	
C	1.5, 2.0, 2.5
D	
E	

TABLE A52. (Continued)

PREPARED BY
CHECKED BY
DATE

NASA-L <sub>2</sub> RC
TEST 290
14.5 x 21.75 VSTX
FULL-SPAN PROPULSIVE WING

A-379

BLN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	$\delta_c$	$\delta_{F_w}$	$\delta_{F_c}$	SCV DAW	DOWN WASH	H/C	NPR	$C_{M_w}$	$C_{M_c}$	Q	$\alpha$	$\beta$	REMARKS	TIME STARTED	TIME FINISHED	DATE
80	BWC	1	1	6	1	0°	45°	45°			A	2.5	2.0	1.0	67	0°	0°				2/1
81	BWC	1	1	6	1	0°	45°	45°			A	2.5	.5	.23	285	0°	0°				2/1
82	BWC	1	1	6	1	0°	45°	45°			A	1.0	0	0	30	0°	0°				2/1
83	BWC	1	1	6	2	0°	45°	45°			$\infty$	C	$\infty$	$\infty$	0	0°	0°	STATIC THRUST RUN			2/1
84	BWC	1	1	6	2	0°	45°	45°			$\infty$	1.0	-	-	0		0°	WT TARE	3:30		2/1
85	BWC	1	1	6	2	0°	45°	45°	Y		$\infty$	1.0	0	0	30	B	0°				2/1
86	BWC	1	1	6	2	0°	45°	45°	Y		$\infty$	2.5	.5	.23	28	B	0°				2/1
87	BWC	1	1	6	2	0°	45°	45°	Y		$\infty$	2.4	1.0	.5	13	B	0°				2/1
88	BWC	1	1	6	2	0°	45°	45°	Y		$\infty$	2.5	2.0	1.0	67	B	0°				2/1
89	BWC	1	1	6	2	0°	45°	45°	Y		A	2.5	.5	.24	27	0°	0°				2/1
89	BWC	1	1	6	2	0°	45°	45°	Y		A	2.5	1.0	.5	138	0°	0°	PE 8, 9, 10, 11, 12 will fix run # @ recompute			2/1
90	BWC	1	1	6	2	0°	45°	45°	Y		A	2.5	2.0	1.0	7	0°	0°				2/1
91																		SPARE RUN NO-USE FOR 89 # 2			
92	BWC	1	1	6	2	0°	45°	45°	Y		A	1.0	0	0	30	0°	0°				2/1
93	BWC	1	1	5	2	0°	45°	45°			$\infty$	1.0	-	-	0		0°	WT TARE NG	8:30		2/1
94	BWC	1	1	5	2	0°	45°	45°			$\infty$	1.0	-	-	0		0°	WT TARE			2/1
95	BWC	1	1	5	2	0°	45°	45°			$\infty$	E	$\infty$	$\infty$	0	0°	0°	STATIC RUN (NG fitting leak)			2/1
96	BVIC	1	1	5	2	0°	45°	45°			$\infty$	E	$\infty$	$\infty$	0	0°	0°	STATIC & WT TARE			2/2
97	BWC	1	1	5	2	0°	45°	45°	Y		$\infty$	1.0	0	0	30	B	0°				2/2
98	BWC	1	1	5	2	0°	45°	45°	Y		$\infty$	2.3	.5	.29	246	B	0°				2/2

NOTES

A
B
C
D
(15, 20, 23

TABLE A52. (Continued)

PREPARED BY
CHECKED BY
DATE

NASA - L <sub>2</sub> RC
TEST 290
14.5 x 21.75 VSTOL
FULL-SPAN PROPULSIVE WING

A-380

RLR NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	$S_c$	$S_{F_w}$	$S_{F_c}$	SCV DATA	DOWN WASH	H/C	NPR	$C_{M_w}$	$C_{M_c}$	Q	$\alpha$	$\beta$	REMARKS	TIME STARTED	TIME ENDED	DATE
99	BWC	1	1	5	2	0°	45°	45°	Y		∞	2.3	1.0	.6	12	B	0°				2/2
100	BWC	1	1	5	2	0°	45°	45°	Y		∞	2.3	2.0	1.2	6.2	B	0°				2/2
101	BWC	1	1	5	2	0°	45°	45°	Y		A	2.3	5	.28	25	0°	0°				2/2
102	BWC	1	1	5	2	0°	45°	45°			A	2.3	1.0	.6	12.5	0°	0°				2/2
103	BWC	1	1	5	2	0°	45°	45°			A	2.3	2.0	1.1	6.3	0°	0°				2/2
104	BWC	1	1	5	2	0°	45°	45°			A	1.0	0	0	30	0°	0°				2/2
105	BWC	1	1	6	3	0°	45°	45°			∞	1.0	-	-	0		0°	WT TARE N.G.			2/2
106	BWC	1	1	6	3	0°	45°	45°			∞	1.0	-	-	0		0°	WT TARE OK			2/2
107	BWC	1	1	6	3	0°	45°	45°			∞	E	∞	∞	0	0°	0°	Static run			2/2
108	BWC	1	1	6	3	0°	45°	45°	Y		∞	1.0	0	0	30	B	0°				2/2
109	BWC	1	1	6	3	0°	45°	45°	Y		∞	2.5	5	21	28	B	0°				2/2
110	BWC	1	1	6	3	0°	45°	45°	Y		∞	2.5	10	42	133	B	0°				2/2
111	BWC	1	1	6	3	0°	45°	45°	Y		∞	2.5	2.0	86	69	B	0°				2/2
112	BWC	1	1	6	3	0°	45°	45°			A	2.4	5	.21	27	0°	0°				2/2
113	BWC	1	1	6	3	0°	45°	45°			A	2.5	1.0	44	13	0°	0°				2/2
113	BWC	1	1	6	3	0°	45°	45°			A	2.5	2.0	85	71	0°	0°	Pe 6, 7, 8, 9, 10			2/2
114	BWC	1	1	6	3	0°	45°	45°			A	1.0	0	0	30	0°	0°				2/2
115																		SPARE RUN No - Use for 113 # 2			
116	BWC	1	1	6	1	10°	45°	45°			∞	1.0	-	-	0		0°	WT TARE			2/2
117	BWC	1	1	6	1	10°	45°	45°			∞	E	∞	∞	0	0°	0°	Static run			2/2

NOTES

A	
B	
C	
D	
E	

TABLE A52. (Continued)

PREPARED BY
CHECKED BY
DATE

NASA-LaRC
TEST 290
14.5 x 21.75 VSTOL
FULL-SPAN PROPULSIVE WING

ALN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	$\delta_c$	$\delta_{FW}$	$\delta_{FC}$	SCV DATA	DOWN WASH	H/C	NPR	$C_{M W}$	$C_{M C}$	Q	$\alpha$	$\beta$	REMARKS	RUN STARTED	RUN FINISHED	DATE
118	BWC	1	1	6	1	10°	45°	45°	Y		∞	1.0	0	0	30	B	0°				2/2
119	BWC	1	1	6	1	10°	45°	45°	Y		∞	2.5	5	.21	27.5	B	0°				2/2
120	BWC	1	1	6	1	10°	45°	45°	Y		∞	2.5	1.0	.43	13.9	B	0°				2/2
121	BWC																	RUN ABORTED - 1 Primly BAD Pt			2/2
122	BWC	1	1	6	1	10°	45°	45°			∞	1.0	-	-	0		0°	WT TARE (New Tape)			2/3
123	BWC								Y									N.G.			
124	BWC	1	1	6	1	10°	45°	45°	Y		∞	2.5	2.0	.88	6.7	B	0°				2/3
125	BWC	1	1	6	1	10°	45°	45°			A	2.5	2.0	.87	7.0	0°	0°				2/3
126	BWC	1	1	6	1	10°	45°	45°			A	2.5	1.0	.44	14	0°	0°				2/3
127	BWC	1	1	6	1	10°	45°	45°			A	2.5	.5	.22	28.5	0°	0°				2/3
128	BWC	1	1	6	1	10°	45°	45°	Y		A	1.0	0	0	30	0°	0°				2/3
129									Y									RUN N.G.			
130	BWC	1	1	6	1	0°	45°	45°			∞	2.4	1.0	.44	12.8	B	0°				2/3
131	BWC	1	1	6	1	0°	45°	45°			∞	2.5	3.0	1.28	4.5	B	0°				2/3
132	BWC	1	1	6	1	0°	45°	45°			∞	2.5	4.0	1.74	3.4	B	0°				2/3
133	BWC	1	1	6	1	0°	45°	45°			∞	2.5	2.0	.87	6.9	2°	A				2/3
134	BWC	1	1	6	1	0°	45°	45°			∞	2.5	.5	.21	2.8	2°	A				2/3
135	BWC	1	1	6	1	0°	45°	45°			∞	1.0	0	0	30	2°	A				2/3
136	BWC	1	1	6	1	0°	45°	45°			B	1.0	0	0	30	2°	0.8				2/3
137	BWC	1	1	6	1	0°	45°	45°			B	2.5	.5	.22	2.8	2°	0.8				2/3

NOTES

A	-0.8 → +0.8
B	-8.0 → +8.0
C	
D	
E	

A-381

TABLE A52. (Continued)

PREPARED BY
CHECKED BY
DATE

NASA-L <sub>2</sub> RC
TEST 290
14 5x 21.75 VSTM
FULL-SPAN PROPULSIVE WING

RLR ID	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	$S_c$	$S_{F_w}$	$S_{F_c}$	SCV DATA	DOWN WASH	H/C	N <sub>PR</sub>	C <sub>M W</sub>	C <sub>M c</sub>	Q	$\alpha$	$\beta$	REMARKS	TIME STARTED	TIME FINISHED	DATE
138	BWC	1	1	6	1	0°	45°	45°			B	2.5	1.0	.44	137	2°	0.8				2/3
139	BWC	1	1	6	1	0°	45°	45°			B	2.5	2.0	.87	69	2°	0.8				2/3
140	BW	1	-	6	-	-	45°	-			$\infty$	1.0	-	0	0°	0°		WT TARE			2/3
141	BW	1	-	6	-	-	45°	-			$\infty$	E	$\infty$	$\infty$	0	0°	0.8	Static Run			2/3
142	BW	1	-	6	-	-	45°	-	Y		$\infty$	1.0	0	0	30	B	0°				2/3
143	BW	1	-	6	-	-	45°	-	Y		$\infty$	2.5	5	0	286	B	0°	10000+ INLB PM			2/3
144	BW	1	-	6	-	-	45°	-	Y		$\infty$	2.5	1.0	0	136	B	0°				2/3
145	BW	1	-	6	-	-	45°	-	Y		$\infty$	2.5	2.0	0	69	B	0°				2/3
146	BW	1	-	6	-	-	45°	-	Y		$\infty$	2.5	3.0	0	49	B	0°	High C <sub>M</sub>			2/3
147	BW	1	-	6	-	-	45°	-	Y		$\infty$	2.5	4.0	0	36	B	0°	" "			2/3
148	BW	1	-	6	-	-	45°	-	Y		$\infty$	2.1	2.0	0	53	B	0°	Constant Q, VARY NPR			2/3
149	BW	1	-	6	-	-	45°	-	Y		$\infty$	1.3	1.0	0	53	B	0°	" " " "			2/3
150	BW	1	-	6	-	-	45°	-	Y		$\infty$	1.0	0	0	30	0°	A		10 55		2/3
151	BW	1	-	6	-	-	45°	-	Y		$\infty$	2.5	.5	0	29.5	0°	A	PM > 10K INLB	11 15		2/3
152	BW	1	-	6	-	-	45°	-	Y		$\infty$	2.5	2.0	0	69	0°	A				2/4
153	BW	1	-	6	-	-	45°	-	Y		A	2.5	2.0	0	6.8	0°	0.8				2/4
154	BW	1	-	6	-	-	45°	-	Y		A	2.5	.5	0	28	0°	0.8	PM > 10K INLB			2/4
155	BW	1	-	6	-	-	45°	-	Y		A	1.0	0	0	30	0°	0.8				2/4
156	BW	1	-	6	-	-	45°	-	Y		A	1.0	0	0	30	0°	0°				2/4
157	BW	1	-	6	-	-	45°	-	Y		A	2.5	.5	0	27.8	0°	0°	PM > 10K INLB	10 10		2/4

NOTES

A
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C
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A-382



TABLE A52. (Continued)

PREPARED BY
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DATE

	NASA - L <sub>o</sub> RC TEST 290 14.5 x 21.75 VSTOL FULL-SPAN PROPULSIVE WING
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A-383

BLN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	$\delta_c$	$\delta_{F_w}$	$\delta_{F_c}$	SCV DATA	DOWN WASH	H/C	NPR	$C_{M_w}$	$C_{M_c}$	Q	$\alpha$	$\beta$	REMARKS	TIME STARTED	TIME FINISHED	DATE
158	BW	1	1	6	-	-	45°	-	Y		A	2.5	1.0	0	13.4	0°	0°		10 25		2/4
159	BW	1	1	6	-	-	45°	-	Y		A	2.5	2.0	0	6.8	0°	0°		10 35		2/4
160	BW	1	-	6	-	-	45°	-	Y		C	2.5	2.0	0	6.9	4°	0°				2/4
161	BW	1	-	6	-	-	45°	-	Y		C	2.5	1.0	0	13.6	4°	0°				2/4
162	BW	1	1	6	-	-	45°	-	Y		C	2.5	.5	0	27.4	4°	0°	PM > 10K INLB			2/4
163	BW	1	-	6	-	-	45°	-	Y		C	1.0	0	0	3.0	4°	0°				2/4
164	BW	1	-	6	-	-	45°	-	Y		$\infty$	2.5	.5	0	28.4	B	0°	Repeat 143 - 10000+ in lb PM			2/4
165	BW	1	-	6	-	-	45°	-	Y		$\infty$	1.9	.5	0	17.6	B	0°	" 164 - lower NPR - 9600 in lb PM			2/4
166	BW	1	-	6	-	-	45°	-	Y		$\infty$	1.0	0	0	A	8°	0°				2/4
167	BW	1	-	6	-	-	45°	-	Y		$\infty$	C	$\infty$	0	0	0	0	WT TARE & STATIC			2/4
168	BW	1	-	6	-	-	45°	-	Y		$\infty$	VARY	.5	0	VARY	8°	0°	10000+ INLB PM			2/4
169	BC	-	1	-	1	10°	-	45°			$\infty$	1.0	-	-	0		0°	WT TARE			2/6
170	BC	-	1	-	1	10°	-	45°			$\infty$	E	0	$\infty$	0	0°	0°	STATIC RUN			2/6
171	BC	-	1	-	1	10°	-	45°	Y		$\infty$	1.0	0	0	3.0	B	0°		9 05		2/6
172	BC	-	1	-	1	10°	-	45°	Y		$\infty$	2.6	0	.25	26.7	B	0°		9 55		2/6
173	BC	-	1	-	1	10°	-	45°	Y		$\infty$	2.6	0	.5	12.7	B	0°		10 25		2/6
174	BC	-	1	-	1	10°	-	45°	Y		$\infty$	2.6	0	1.0	6.8	B	0°			11 00	2/6
175	BC	-	1	-	1	0°	-	45°			$\infty$	E	0	$\infty$	0	0°	0°	STATIC RUN	11 30		2/6
176	BC	-	1	-	1	0°	-	45°	Y		$\infty$	1.0	0	0	3.0	B	0°		12 10		2/6
177	BC	-	1	-	1	0°	-	45°	Y		$\infty$	2.5	0	.25	25.6	B	0°		1 25		2/6

NOTES

A	
B	
C	
D	
E	

TABLE A52. (Continued)

PREPARED BY
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DATE

NASA-L <sub>o</sub> RC
TEST 290
145 x 21.75 VSTM
FULL-SPAN PROPULSIVE WING

BLR ID	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	SC	S <sub>FW</sub>	S <sub>FC</sub>	SCV DATA	DOWN WASH	H/C	NPR	C <sub>M W</sub>	C <sub>M C</sub>	Q	α	β	REMARKS	TIME STARTED	TIME FINISHED	DATE
178	BC	-	1	-	1	0°	-	45°	Y		∞	25	0	.5	128	B	0°		1:50		2/6
179	BC	-	1	-	1	0°	-	45°	Y		∞	25	0	10	60	B	0°		2:10		2/6
180	B	-	-	-	-	-	-	-			∞	1.0	-	-	0		0°	WT TARE	2:55		2/6
181	B	-	-	-	-	-	-	-	Y		∞	1.0	0	0	30	B	0°		3:15		2/6
182	BW	1/2	-	6	-	-	45°	-			∞	VARY	∞	0	0	0°	0°	PT <sub>NO2</sub> VS PPL calib			2/6
183	BW	1/2	-	6	-	-	45°	-			∞	E	∞	0	0	0°	0°	WT TARE + STATIC RUN			2/6
184	BW	1/2	-	6	-	-	45°	-			∞	E	∞	0	0	0°	0°	STATIC RUN			2/6
185	BW	1/2	-	6	-	-	45°	-			∞	VARY	∞	0	0	0°	0°	PT <sub>NO2</sub> VS PPL <sub>EN</sub> CALIB			2/6
186	BW	1/2	-	6	-	-	45°	-			∞	F	∞	0	0	0°	0°	STATIC RUN			2/6
187	BW	1/2	-	6	-	-	45°	-	Y		∞	1.0	0	0	30	B	0°		11:00		2/6
188	BW	1/2	-	6	-	-	45°	-	Y		∞	2.5	5	0	28.7	B	0°		11:15		2/6
189	BW	1/2	-	6	-	-	45°	-	Y		∞	2.5	1	0	149	B	0°		11:25		2/6
190	BW	1/2	-	6	-	-	45°	-	Y		∞	2.5	2	0	7.5	B	0°		11:40		2/6
191	BW	1/2	-	6	-	-	45°	-	Y		∞	1.0	0	0	30	0°	B		10:15		2/7
192	BW	1/2	-	6	-	-	45°	-	Y		∞	2.5	.5	0	305	0°	B		11:00		2/7
193	BW	1/2	-	6	-	-	45°	-	Y		∞	2.5	2	0	7.6	0°	B				
194	BW	1/2	-	6	-	-	45°	-			∞	1.0	-	-	0		0°	WT TARE (w/ 40 lb WTs)			2/7
195	BW	1/2	-	6	-	-	45°	-	Y		∞	G	∞	0	0	0°	0°	STATIC RUN			2/7
196	BW	1/2	-	6	-	-	45°	-			∞	1.0	-	-	0		0°	WT TARE (w/o 40 lb WTs)	3:05		2/7
197	BW	1/2	-	6	-	-	45°	-			∞	G	∞	0	0	0°	0°	STATIC RUN			2/7

A-384

NOTES

A	
B	0.0 → +8.0
C	
D	
E	

TABLE A52. (Continued)

PREPARED BY
CHECKED BY
DATE

NASA-LARC
TEST 290
14.5 x 21.75 VSTM
FULL-SPAN PROPULSIVE WING

A-385

RUN NO	CONFIGURATION	W. NOZ SPAN	C. NOZ SPAN	WING POS	CAZ POS	S <sub>C</sub>	S <sub>FW</sub>	S <sub>FC</sub>	SCV DATA	DOWN WASH	H/C	NPR	C <sub>M W</sub>	C <sub>M C</sub>	Q	α	β	REMARKS	TIME STARTED	TIME FINISHED	DATE
198	BW	1/2	-	6	-	-	45°	-										BAL INTERFERENCE CHK RUN			2/7
199	BW	1/2	-	6	-	-	45°	-										" " " "			2/7
200	BW	1/2	-	6	-	-	45°	-			∞	1.0	0	0	30	B	0°				2/7
201	BWC	1/2	1	6	1	0°	45°	45°			∞	1.0	-	-	0		0°	WT TARE	6:45		2/7
202	BWC	1/2	1	6	1	0°	45°	45°			∞	VARY	∞	∞	0	0°	0°	STATIC RUN	7:20		2/7
203	BWC	1/2	1	6	1	0°	45°	45°			∞	VARY	∞	∞	0	0°	0°	" "			2/7
204																		No Data wt tare??			2/7
205	B																	STING Press Tare Run			2/7
206																		Model loading run w/ Jasic			2/8
207																		" " " "			2/8
208																		" " " "			2/8
209																		" " " "			2/8
210																		" " " "			2/8
211	B																	STING PRESS TARE RUN			2/8
212	BWC	1/2	1	6	1	0°	45°	45°			∞	VARY	∞	∞	0	0°	0°	STATIC THRUST RUN NG			2/8
213	BWC	1/2	1	6	1	0°	45°	45°			∞	VARY	∞	∞	0	0°	0°	" " " "			2/8
214	BWC	1/2	1	6	1	0°	45°	45°			∞	VARY	∞	∞	0	20°	0°	" " " α=20°			2/8
215	BWC	1/2	1	6	1	0°	45°	45°			∞	1.97	.5	.31	17	B	0°		4:35		2/8
216	BWC	1/2	1	6	1	0°	45°	45°	Y		∞	1.0	0	0	20	B	0°		5:15		2/8
217	BWC	1/2	1	6	1	0°	45°	45°	Y		∞	1.92	1.0	.63	79	B	0°		5:40		2/8

NOTES

A
B
C
D
E

TABLE A52. (Continued)

PREPARED BY
CHECKED BY
DATE

NASA - L <sub>o</sub> RC
TEST 290
14.5 x 21.75 VSTOL
FULL-SPAN PROPULSIVE WING

RLN NO	CONFIGURATION	1/2 NOZ SPAN	C-NOZ SPAN	WING POS	CAZ POS	SC	S <sub>FW</sub>	S <sub>FC</sub>	SCV DATA	DOWN WASH	H/C	NPR	C <sub>M W</sub>	C <sub>M C</sub>	Q	α	β	REMARKS	TIME STARTED	TIME FINISHED	DATE
218	BWC	1/2	1	6	1	0°	45°	45°	Y		∞	1.93	2.0	1.22	41	B	0°				2/8
219	BW	1/2	-	6	-	-	45°	-			∞	1.0	-	-	0		0°	WT TARE			2/8
220	BW	1/2	-	6	-	-	45°	-			∞	1.6 2.3	∞	0	0	0°	0°	STATIC RUN			2/8
221	BW	1/2	-	6	-	-	45°	-	Y		∞	1.0	0	0	20	B	0°				2/8
222	BW	1/2	-	6	-	-	45°	-			∞	1.7	.5	0	14.4	E	0°		7:10		2/8
223	BW	1/2	-	6	-	-	45°	-			∞	1.7	1.0	0	8.1		0°				2/8
224	BW	1/2	-	6	-	-	45°	-			∞	1.9	1.0	0	9		0°				2/8
225	BW	1/2	-	6	-	-	45°	-			∞	1.9	2.0	0	5.1		0°				2/8
226	BW	1/2	-	6	-	-	45°	-	Y		∞	1.95	2.0	0	5		A				2/8
227	BW	1/2	-	6	-	-	45°	-	Y		A	1.9	2.0	0	4.7	0°	0°				2/8
228	BW	1/2	-	6	-	-	45°	-	Y		A	1.9	1.0	0	9	0°	0°				2/8
229	BW	1/2	-	6	-	-	45°	-	Y		A	1.7	.5	0	16.7	0°	0°				2/8
230	BW	1/2	-	6	-	-	45°	-	Y		A	1.6	0	0	20	0°	0°				2/8
231	BW	1/2	-	6	-	-	45°	-	Y		D	1.0	0	0	20	8°	0°				2/8
232	BW	1/2	-	6	-	-	45°	-	Y		D	1.7	.5	0	16.5	8°	0°				2/8
233	BW	1/2	-	6	-	-	45°	-	Y		D	1.7	1.0	0	7.8	8°	0°	Pt 1,2,3			2/8
233	BW	1/2	-	6	-	-	45°	-	Y		D	1.7	2.0	0	4.2	8°	0°	Pt 4,7,9			2/8
234	BW	1/2	-	6	-	-	45°	-			D	1.7	.74	0	11.6	8°	0°		11:05		2/8
235	BW	1/2	-	6	-	-	45°	-	Y		∞	1.7	.5		16	0°	B				2/8
236	BW	1/2	-	6	-	-	45°	-	Y		∞	1.0	0		20	0°	B				2/8

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TABLE A52. (Continued)

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NASA-LARC
TEST 290
14.5 x 21.75 VSTOL
FULL-SPAN PROPULSIVE WING

A-387

Run No	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	$S_c$	$S_{F_w}$	$S_{F_c}$	SCV DATA	DOWN WASH	H/C	NPR	$C_{M_w}$	$C_{M_c}$	Q	$\alpha$	$\beta$	REMARKS	TIME STARTED	TIME FINISHED	DATE
237	BWC	1/2	1	6	1	0°	0°	0°			∞	10	-	-	0		0°	WT TARE	9.30		2/9
238	BWC	1/2	1	6	1	0°	0°	0°			∞	10	-	-	0		0°	WT TARE	9.45		2/9
239	BWC	1/2	1	6	1	0°	0°	0°			∞	VARY	∞	∞	0		0°	STATIC RUN	10.00		2/9
240	BWC	1/2	1	6	1	0°	0°	0°	Y		∞	10	0	0	30	A	0°		10.45		2/9
241	BWC	1/2	1	6	1	0°	0°	0°	Y		∞	2.0	.5	.31	21.1	B	0°		11.15		2/9
242	BWC	1/2	1	6	1	0°	0°	0°			∞	2.0	1.0	.66	10.6	B	0°		11.57		2/9
																		CHANGE TO BENT STING			
243	B																	STING-PRESS TAKE RUN			2/9
244																		? POSSIBLY XDU CER CALIB			2/10
245																		? POSSIBLY Hgt & $\alpha$ CALIB			2/10
246																		? PLEN PRESS "			2/10
247	B																	STING-PRESS TARE RUN			2/10
248	BWC	1/2	1	6	1	0°	0°	0°			∞	VARY	∞	∞	0	0°	0°	STATIC RUN	11.05		2/10
249	BWC	1/2	1	6	1	0°	0°	0°			∞	1.0	-	-	0		0°	WT TARE	11.35		2/10
250	BWC	1/2	1	6	1	0°	0°	0°			∞	2.0	.5	.28	17.5	B	0°		12.55		2/10
251	BWC	1/2	1	6	1	0°	0°	0°			∞	2.4	1.0	.54	12	B	0°				2/10
252	BWC	1/2	1	6	1	0°	0°	0°			∞	2.4	2.0	1.1	6	B	0°				2/10
253	BW	1/2	-	6	-	-	0°	-			∞	VARY	∞	-	0	0°	0°	Wing P <sub>TNOZ</sub> vs P <sub>PLEN</sub> CALIB			2/10
254	BW	1/2	-	6	-	-	0°	-			∞	1.0	-	-	0		0°	WT TARE			2/10
255	BW	1/2	-	6	-	-	0°	-			∞	1.5 2.4	∞	0	0	0	0°	STATIC RUN			2/10

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TABLE A52. (Continued)

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NASA-LARC
TEST 290
14 5 x 21.75 VSTM
FULL-SPAN PROPULSIVE WING

A-388

Run No	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	$\delta_c$	$S_{FW}$	$S_{FC}$	SCV DATA	DOWN WASH	H/C	NPR	$C_{M W}$	$C_{M C}$	Q	$\alpha$	$\beta$	REMARKS	TIME STARTED	TIME FINISHED	DATE
256	BW	1/2	-	6	-	-	0°	-	Y		∞	1.0	0	0	30	B	0°				2/10
257	BW	1/2	-	6	-	-	0°	-	Y		∞	2.4	.5	0	28	B	0°		5:45		2/10
258	BW	1/2	-	6	-	-	0°	-	Y		∞	2.4	1.0	0	14	B	0°				2/10
259	BW	1/2	-	6	-	-	0°	-	Y		∞	2.4	2.0	0	8	B	0°				2/10
260	BW	1/4	-	6	-	-	0°	-			∞	VARY	∞	0	0	0°	0°	1/4 SPAN NOZZ/PLEN CALIB	10:13		2/10
261	BW	1/4	-	6	-	-	0°	-			∞	VAR	∞	0	0	0°	0°	STATIC RUN (use 262)	10:20		2/10
262	BW	1/4	-	6	-	-	0°	-	Y		∞	1.0	0	0	30	B	0°		10:40		2/10
263	BW	1/4	-	6	-	-	0°	-			∞	1.0	-	-	0		0°	WT TARE (NEW TAPE)			2/11
264	BW																	Vent AP Xducer Calib	8:45		2/11
265	BW	1/4	-	6	-	-	0°	-	Y		∞	2.4	.5	0	174	B	0°		9:30		2/11
266	BW	1/4	-	6	-	-	0°	-	Y		∞	2.4	1.0	0	79	B	0°				2/11
267	BW	1/4	-	6	-	-	0°	-	Y		∞	2.4	2.0	0	41	B	0°				2/11
268	BW	1/4	-	6	-	-	0°	-			∞	2.2 <sup>1.5</sup>	∞	0	0	0°	0°	STATIC RUN (C <sub>M</sub> calib)	10:30		2/11
269	BWC	1/4	1	6	1	0°	0°	0°			∞	2.3 <sup>1.5</sup>	∞	∞	0	0°	0°	STATIC RUN	11:15		2/11
270	BWC	1/4	1	6	1	0°	0°	0°			∞	1.0	-	-	0		0°	WT TARE	11:25		2/11
271	BWC	1/4	1	6	1	0°	0°	0°	Y		∞	1.0	0	0	30	B	0°		11:35		2/11
272	BWC	1/4	1	6	1	0°	0°	0°	Y		∞	2.4	.5	.3	18	B	0°		11:55		2/11
273	BWC	1/4	1	6	1	0°	0°	0°	Y		∞	2.4	1.0	.6	9	B	0°		12:15		2/11
274	BWC	1/4	1	6	1	0°	0°	0°	Y		∞	2.4	2.0	1.2	45	B	0°		12:30		2/11
275	BW	1/4	-	6	-	-	45°	-			∞	2.3 <sup>1.5</sup>	∞	∞	0	0°	0°	Static run NG	2:40		2/11

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TABLE A52. (Continued)

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NASA-LaRC
TEST 290
14.5 x 21.75 VSTOL
FULL-SPAN PROPULSIVE WING

A-389

RLR NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	S <sub>c</sub>	S <sub>F<sub>w</sub></sub>	S <sub>F<sub>c</sub></sub>	SCV DATA	DOWN WASH	H/C	N <sub>PR</sub>	C <sub>M W</sub>	C <sub>M C</sub>	Q	α	β	REMARKS	TIME STARTED	TIME FINISHED	DATE
276	BW	1/4	-	6	-	-	45°	-			∞	1.0	-	-	0		0°	WT TARE	2:45		2/11
277	BW	1/4	-	6	-	-	45°	-	Y		∞	1.0	0	0	30	B	0		3:00		2/11
278	BW	1/4	-	6	-	-	45°	-			∞	VARY	∞	0	0	0°	0°	STATIC RUN N.G.	4:00		2/11
279	BW	1/4	-	6	-	-	45°	-			∞	VARY	∞	0	0	0	0	STATIC RUN			2/11
280	BW	1/4	-	6	-	-	45°	-	Y		∞	1.9	.5	0	10.4	B	0°				2/11
281	BW	1/4	-	6	-	-	45°	-	Y		∞	1.9	1.0	0	4.7	B	0°				2/11
282	BW	1/4	-	6	-	-	45°	-	Y		∞	1.9	2.0	0	2.7	B	0°				2/11
283	BW	1/4	-	6	-	-	45°	-	Y		∞	1.0	0	0	30	0°	B		6:20		2/11
284	BW	1/4	-	6	-	-	45°	-	Y		∞	1.9	.5	0	10	0°	B				2/11
285	BW	1/4	-	6	-	-	45°	-	Y		∞	1.9	2	0	2.6	0°	B				2/11
286	BWC	1/4	1	6	1	0°	45°	45°			∞	1.0	-	-	0		0	WT TARE N.G.	8:00		2/11
287	BWC	1/4	1	6	1	0°	45°	45°			∞	1.0	-	-	0		0	WT TARE			2/11
288	BWC	1/4	1	6	1	0°	45°	45°			∞	VARY	∞	∞	0	0°	0°	STATIC RUN			2/11
289	BWC	1/4	1	6	1	0°	45°	45°	Y		∞	1.0	0	0	30	B	0°				2/11
290	BWC	1/4	1	6	1	0°	45°	45°	Y		∞	1.9	.5	.27	13.5	B	0°				2/11
291	BWC	1/4	1	6	1	0°	45°	45°	Y		∞	1.95	1.0	.55	6.4	B	0°				2/11
292	BWC	1/4	1	6	1	0°	45°	45°	Y		∞	1.96	2.0	1.1	3.4	B	0°				2/11
293	BWC	1/4	1/2	6	1	0°	45°	45°			∞	VARY	∞	∞	0	0°	0°	Nozz Calib			
294	BWC	1/4	1/2	6	1	0°	45°	45°			∞	1.0	-	-	0		0°	WT TARE			
295	BWC	1/4	1/2	6	1	0°	45°	45°			∞	VARY	∞	∞	0	0°	0°	STATIC RUN			

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TABLE A52. (Concluded)

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NASA-L-RC
TEST 290
145x2175 VSTOL
FULL-SPAN PROPULSIVE WING

A-390

RLN ID	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	SC	S <sub>F<sub>w</sub></sub>	S <sub>F<sub>c</sub></sub>	SCV DATA	DOWN WASH	H/C	NPR	C <sub>M w</sub>	C <sub>M c</sub>	Q	α	β	REMARKS	TIME STARTED	TIME FINISHED	DATE
296	BWC	1/4	1/2	6	1	0°	45°	45°	Y		∞	1.95	5	.3	12.4	B	0°			3/12	
297	BWC	1/4	1/2	6	1	0°	45°	45°	Y		∞	1.95	10	.66	6.1	B	0°			3/12	
298	BWC	1/4	1/2	6	1	0°	45°	45°	Y		∞	1.96	2.1	1.27	3.2	B	0°			3/12	
299	BC	-	1/2	-	1	0°	-	45°			∞	1.0	-	-	0		0°	WT TARE		3/12	
300	BC	-	1/2	-	1	0°	-	45°			∞	VARY	0	∞	0	0°	0°	STATIC RUN		3/12	
301	BC	-	1/2	-	1	0°	-	45°			∞	1.96	0	28	18.7	B	0°	PTS 1-8		3/12	
301	BC	-	1/2	-	1	0°	-	45°			∞	1.95	0	59	10.0	B	0°	PTS 9-15 (1 PT @ C <sub>M<sub>c</sub></sub> = .92)		3/12	
303	BC	-	1/2	-	1	0°	-	45°			∞	1.94	0	1.09	4.5	B	0°			3/12	
304	BW	1	-	6	-	-	45°	-			∞	1.5 2.4	∞	0	0	0	0°	PT102/PALLEN Calib		3/12	
305	BW	1	-	6	-	-	45°	-			∞	1.0	-	-	0		0°	WT Tare N.G		3/12	
306	BW	1	-	6	-	-	45°	-			∞	1.0	-	-	0		0°	WT Tare OK		3/12	
307	BW	1	-	6	-	-	45°	-			∞	1.5 2.2	∞	0	0	0	0°	Static Run		3/12	
308	BW	1	-	6	-	-	45°	-	151		∞	1.8	5	0	20	0	B			3/12	
309	BW	1	-	6	-	-	45°	-	-		∞	1.8	1.0	0	9.7	0	B			3/12	
310	BW	1	-	6	-	-	45°	-	152		∞	1.8	2.0	0	5	0	B			3/12	
311	BW	1	-	6	-	-	45°	-	145 148		∞	1.9	2.0	0	5	B	0°	See 145, 148		3/12	
312	BW	1	-	6	-	-	45°	-	144 149		∞	1.8	1.1	0	9	B	0°	See 144, 149. For SCV		3/12	
313	BW	1	-	6	-	-	45°	-	145		∞	1.8	5	0	20	C	0°			3/12	

NOTES

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16 Abstract A full span model of a wing/canard concept representing a fighter configuration has been tested at STOL conditions in the NASA Langley 4 x 7 meter tunnel. The results of this test are presented, and comparisons are made to previous data of the same configuration tested as a semispan model. The potential of the propulsive wing/canard to develop very high lift coefficients was investigated with several nozzle spans (nozzle aspect ratios). Although longitudinal trim was not accomplished with the blowing distributions and configurations tested, the propulsive wing/canard appears to offer an approach to managing the large negative pitching moments associated with trailing edge flap blowing. Also presented are data showing the effects of large flap deflections and relative wing/canard positions. Presented in the appendix to the report are limited lateral-directional and ground effects data, as well as wing downwash measurements.					
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