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TABULATED PRESSURE MEASUREMENTS ON AN NASA SUPERCRITICAL-WING RESEARCH AIRPLANE MODEL WITH AND WITHOUT FUSELAGE AREA-RULE ADDITIONS AT MACH 0.25 TO 1.00

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TABULATED PRESSURE MEASUREMENTS ON AN NASA SUPERCRITICAL-WING RESEARCH AIRPLANE MODEL WITH AND WITHOUT FUSELAGE AREA-RULE ADDITIONS AT MACH 0.25 TO 1.00*

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SUMMARY

In order to determine the effects of side fuselage area-rule additions on the local aerodynamic loads over the wing and rear fuselage of a supercritical-wing research airplane, basic pressure measurements have been made on a 0.087-scale model in the Langley 8-foot transonic pressure tunnel at Mach numbers from 0.25 to 1.00. In addition, pressure measurements over the surface of the area-rule additions themselves were obtained at angles of sideslip of approximately -5°, 0°, and 5° to aid in the structural design of the additions. Except for representative figures, results are presented in tabular form without analysis.

INTRODUCTION

A new airfoil developed by the National Aeronautics and Space Administration (refs. 1 to 7) has demonstrated potential for permitting increases in wing thicknesses on subsonic aircraft without incurring reductions in drag divergence Mach number or, conversely, increases in drag divergence Mach number relative to conventional airfoils of comparable thicknesses. A wide range of wind-tunnel investigations (refs. 7 to 11, for example) of several airplane model configurations incorporating the supercritical-airfoil concept have indicated improvements in performance and maneuver capabilities with marked potential for both military and commercial applications of the supercritical airfoil. Typical applications are considered in references 12 and 13.

Initial full-scale flight tests are currently being conducted (ref. 7) on a supercritical wing (planform typical of that of proposed advanced technology transport configurations) mounted on a test-bed airplane to provide flight verification of wind-tunnel data at Mach numbers through 1.00. An area-rule follow-on configuration to be flown during later phases of the flight test program is under development. This follow-on configuration incorporates experimentally developed fore and aft area-rule additions to the sides

*Title, Unclassified.

of the fuselage to improve the longitudinal development of cross-sectional area and is more representative of future transport application. These area-rule additions and their effects on the total longitudinal aerodynamic force and moment characteristics of the supercritical-wing research airplane model are discussed in reference 14. Selected pressure measurements at near-cruise conditions are also presented in reference 14.

This report documents the aerodynamic load distributions over the wing and rear fuselage of a 0.087-scale model with and without the fuselage area-rule additions at Mach numbers from 0.25 to 1.00. Pressure measurements over the surface of the area-rule additions at sideslip angles of approximately -5°, 0°, and 5° are also included. Except for representative figures, results are presented in tabular form without analysis.

SYMBOLS

Values are given in both SI and U.S. Customary Units. Measurements and calculations were made in U.S. Customary Units. Aerodynamic coefficients are based on the dimensions of the basic (reference) wing panel which does not include the leading-edge glove or the trailing-edge extension (fig. 1(a)). Wing-section pitching-moment coefficients are referenced to the local 25-percent-chord line of the basic wing panel.

The pressure data presented herein were tabulated by machine, and the limitations of the machine as to available type faces necessitated some differences between the notation of these tables and conventional symbols. The symbols are given in the conventional form with the machine notation included in parentheses.

b — wing span, 114.30 cm (45.00 in.)

C_L — total lift coefficient, Lift/qS

C_p (CP) — pressure coefficient, $\dfrac{p_l - p}{q}$

c — local streamwise chord of basic wing panel

c_m (CM) — wing-section pitching-moment coefficient about 0.25c,
$$\int_{l.e.}^{t.e.} \left(C_{p,L} - C_{p,U}\right)\left(0.25 - \frac{x}{c}\right) d\left(\frac{x}{c}\right)$$

c_n (CN) — wing-section normal-force coefficient, $\int_{l.e.}^{t.e.} \left(C_{p,L} - C_{p,U}\right) d\left(\frac{x}{c}\right)$

c' — local streamwise chord of total wing planform which includes leading-edge glove and trailing-edge extension

M free-stream Mach number

p free-stream static pressure

p_l local static pressure

q free-stream dynamic pressure

S area of basic wing panels including fuselage intercept, 0.193 m^2 (2.075 ft^2)

x/c (X/C) longitudinal location of pressure orifice where x is distance in cm (in.) from leading edge of local chord of basic wing panel

x' streamwise distance measured from leading edge of total wing planform

y spanwise distance measured normal to plane of symmetry

z' vertical distance measured from model reference water line 26.205 cm (10.317 in.)

α angle of attack referred to a model water line, deg

β angle of sideslip referred to model center line, deg

δ_h horizontal-tail deflection angle referred to a model water line, positive when trailing edge down, deg

θ circumferential location of pressure orifices on rear of fuselage, deg

Subscripts:

L wing lower surface

U wing upper surface

Abbreviations:

l.e. leading edge

t.e. trailing edge

APPARATUS AND PROCEDURES

Model

Geometric characteristics of the sting-supported 0.087-scale pressure model are presented in figure 1 and photographs of the model are presented as figure 2. Figure 3 shows the model geometric cross-sectional area progression with and without the fuselage area-rule additions. Wing-section coordinates were identical to those of the wings used in the investigations reported in references 8, 9, and 14 and are repeated in table I. Ratios of streamwise thickness to total chord varied from about 12 percent at the wing root to 9 percent at the mean aerodynamic chord (fig. 1(a)) and to about 7 percent at the wing tip. The wing had a root incidence angle of 1.5° and approximately 5° of twist (washout) from root to tip in the unloaded condition.

The wing was constructed with steel leading and trailing edges and with a steel core around which plastic fill was used to form the upper and lower surfaces and in which steel pressure tubing was embedded.

The model, with and without the area-rule additions, was identical to the model of reference 14, and the model without the area-rule additions was considered to be a very accurate representation of the full-scale airplane. Major full-scale airplane protuberances simulated on the model are identified in figure 2. The drogue-parachute fairing shown near the model base at the foot of the vertical tail was not included on the model for the present investigation, nor is it on the full-scale airplane. The underwing leading-edge vortex generators are comprehensively discussed in reference 15.

Test Facility

The Langley 8-foot transonic pressure tunnel is a single-return rectangular wind tunnel with controls that permit the independent variation of Mach number, stagnation pressure and temperature, and dewpoint. The upper and lower test-section walls are axially slotted to permit testing through the transonic speed range. The total slot width at the position of the model averaged about 6 percent of the width of the upper and lower walls. In addition to the slots, wooden test-section sidewall inserts (fig. 1(g)) indented in the region of the model were used to reduce blockage effects at Mach numbers near 1.00. (See ref. 15.)

Surface Pressure Measurements

Pressures were measured with the use of electronically actuated differential pressure scanning valves. The orifice locations are diagramed and listed in table II.

Wing.- Pressures were measured along six streamwise rows over the upper surface of the right wing panel and the lower surface of the left wing panel. Attention is called to wing semispan stations $\frac{y}{b/2}$ of 0.307 and 0.480 (table II(a)) where instrumentation problems resulted in relatively fewer lower-surface orifices than for the other wing semispan stations. Wing-section normal-force and pitching-moment coefficients were obtained by numerical integration (based on the trapezoidal method) of the local pressure coefficients measured at each orifice multiplied by an appropriate weighting factor (incremental area).

Rear fuselage.- Four short longitudinal rows of pressure orifices were distributed over the rear of the fuselage, near the base, at $\theta \approx 8^{\circ}$, 46°, 136°, and 180° (table II(b)).

Area-rule additions.- Pressures were measured over the surface of the area-rule additions (table II(c)) on the left side of the fuselage to provide data on the aerodynamic load distribution over the area-rule additions themselves. These measurements are supplemental to the single row of pressure measurements along the side of the fuselage near the crest of the area-rule additions presented in reference 15.

Accuracy and Corrections

Adjustments have been made to the measured angles of attack to account for deflections of the model balance and sting support system under aerodynamic load. The accuracy with which angle of attack may be determined generally decreases with increases in aerodynamic load and model support system dynamics. At the maximum lift coefficients of this investigation, the angles of attack were estimated to be accurate within $\pm 0.1^{\circ}$. At near-cruise conditions, the angles of attack were considered to be accurate within $\pm 0.05^{\circ}$. Further corrections to the measured angles of attack have been made for tunnel airflow angularity (determined from comparisons of results of tests with upright and inverted models).

The ranges of the transducers in the differential pressure scanning valves were ± 103.4 kN/m^2 (± 15 lb/in^2) for the wing upper surface, ± 82.7 kN/m^2 (± 12 lb/in^2) for the wing lower surface, ± 68.9 kN/m^2 (± 10 lb/in^2) for the surface of the area-rule additions, and ± 17.2 kN/m^2 (± 2.5 lb/in^2) for the rear fuselage. Estimated accuracies of these transducers are 1 percent of the maximum ranges.

The Mach number was considered to be accurate within 0.002.

An attempt was made to compensate for the relatively few pressure orifices on the lower surface of semispan stations $\frac{y}{b/2}$ of 0.307 and 0.480 by adjusting the weight (incremental area) given to each orifice during the section normal-force- and pitching-moment-coefficient numerical integration procedure. Such adjustments were made on the basis of wing pressure distributions observed during earlier model testing.

Test Conditions

Tests were conducted at Mach numbers from 0.25 to 1.00 at the specific conditions listed in table III. The stagnation temperature of the tunnel air was automatically controlled at approximately 322 K (120° F) and the air was dried until the dewpoint in the test section was reduced sufficiently to avoid condensation effects (ref. 16).

Both model configurations were tested with fixed boundary-layer transition on the wing as shown in figure 4. Boundary-layer transition trips of No. 120 carborundum grains were located on the horizontal and vertical tails at 5 percent of the local streamwise chord. Transition trips of No. 120 carborundum grains were also fixed around the fuselage 2.54 cm (1.00 in.) behind the model nose and on both the inner and outer surfaces of the flow-through duct inlet 1.27 cm (0.50 in.) rearward of the inlet lip. All transition trips were 0.13 cm (0.05 in.) wide.

SUMMARY OF DATA PRESENTED

Typical pressure distributions over the wing and rear fuselage with and without the area-rule additions at several Mach numbers are presented in figures 5 and 6 at angles of attack near those at which the cruise lift coefficient of 0.40 occurs.

Complete wing pressure profile tabulations along with the integrated wing-section normal-force and pitching-moment coefficients (c_n and c_m) for the model without and with fuselage area-rule additions at a horizontal-tail deflection angle of -2.5° are presented in tables IV and V, respectively. Pressure distributions over the rear fuselage in the vicinity of the horizontal tail with and without the area-rule additions are presented for various horizontal-tail deflection angles in tables VI to IX. Nominal rear fuselage orifice locations are used in tables VI to IX to simplify tabulation. Actual orifice locations which vary only slightly from those used in the tabulations are as shown in table II(b). Pressure distributions over the surface of the area-rule additions are presented in table X for angles of sideslip of approximately -5°, 0°, and 5°.

Langley Research Center,
National Aeronautics and Space Administration,
Hampton, Va., October 5, 1972.

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TABLE I. - WING COORDINATES ALONG STREAMWISE CHORDS

(a) Wing planform coordinate layout

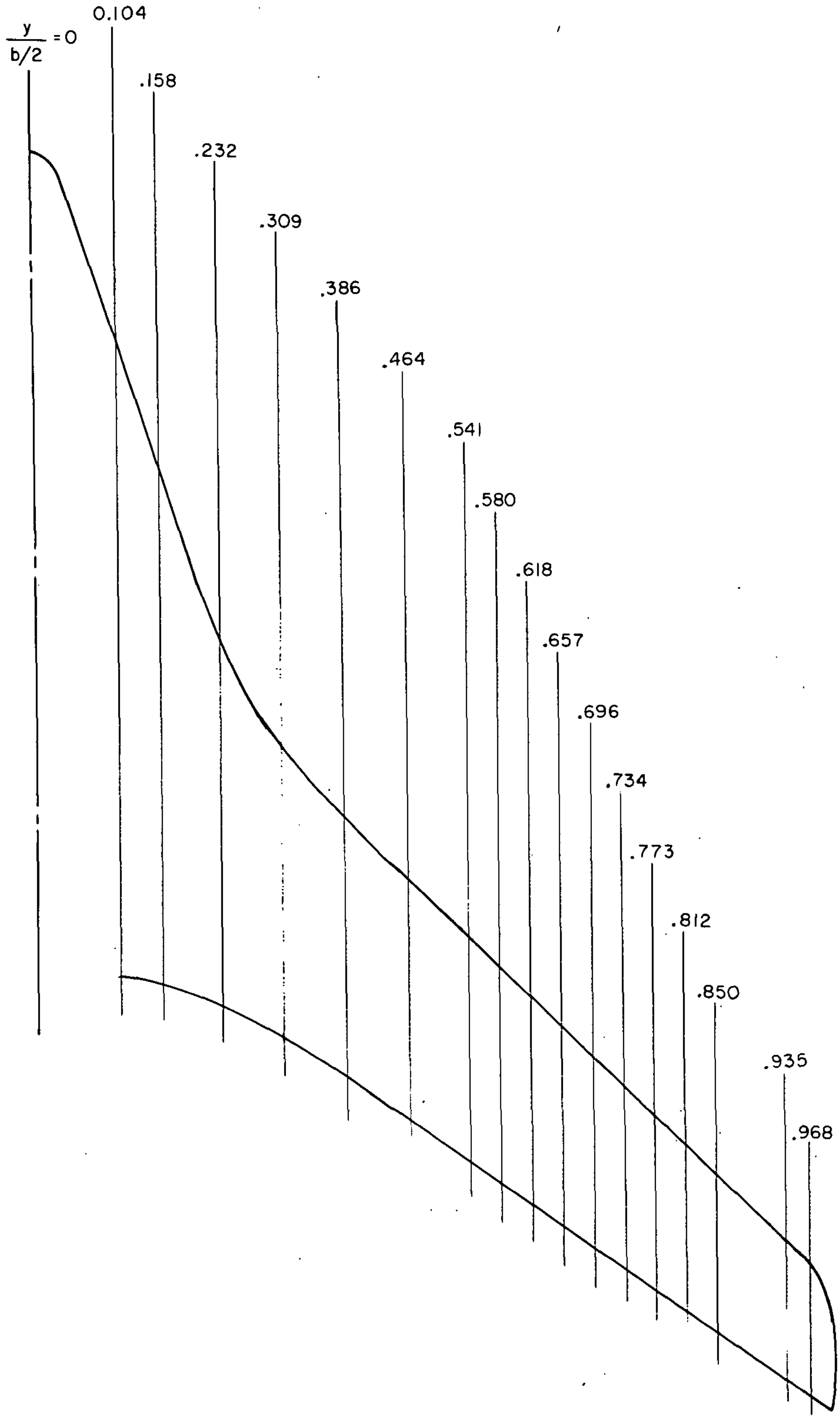

TABLE I.- WING COORDINATES ALONG STREAMWISE CHORDS – Continued

(b) $\frac{y}{b/2} = 0.104$; $c' = 45.839$ cm (18.047 in.)

x'/c'	z'/c' Upper surface	z'/c' Lower surface	x'/c'	z'/c' Upper surface	z'/c' Lower surface	x'/c'	z'/c' Upper surface	z'/c' Lower surface
0	0.0379	0.0379	0.0240	0.0608	0.0133	0.0958	0.0777	-0.0057
.00001	.0384	.0373	.0250	.0613	.0128	.0979	.0781	-.0061
.00004	.0389	.0368	.0261	.0617	.0123	.1000	.0784	-.0064
.00010	.0395	.0362	.0272	.0621	.0118	.1022	.0787	-.0068
.0002	.0400	.0357	.0283	.0625	.0114	.1043	.0790	-.0071
.0003	.0406	.0351	.0294	.0629	.0109	.1065	.0793	-.0075
.0004	.0411	.0345	.0306	.0633	.0104	.1087	.0797	-.0078
.0006	.0417	.0340	.0318	.0637	.0100	.1110	.0800	-.0081
.0007	.0422	.0334	.0330	.0641	.0095	.1132	.0803	-.0085
.0009	.0428	.0329	.0342	.0645	.0091	.1155	.0806	-.0088
.0011	.0433	.0323	.0355	.0649	.0086	.1178	.0809	-.0091
.0014	.0438	.0318	.0368	.0653	.0082	.1201	.0812	-.0094
.0016	.0444	.0312	.0381	.0657	.0078	.1224	.0815	-.0097
.0019	.0449	.0307	.0394	.0661	.0073	.1248	.0818	-.0100
.0022	.0454	.0301	.0408	.0665	.0069	.1272	.0821	-.0103
.0025	.0460	.0296	.0421	.0669	.0065	.1296	.0824	-.0107
.0029	.0465	.0290	.0435	.0673	.0061	.1321	.0827	-.0110
.0033	.0470	.0285	.0449	.0677	.0056	.1345	.0829	-.0113
.0037	.0475	.0279	.0464	.0681	.0052	.1370	.0832	-.0116
.0041	.0481	.0274	.0478	.0685	.0048	.1395	.0835	-.0119
.0045	.0486	.0268	.0493	.0688	.0044	.1420	.0838	-.0122
.0050	.0491	.0263	.0508	.0692	.0040	.1446	.0840	-.0125
.0055	.0496	.0257	.0523	.0696	.0036	.1471	.0843	-.0128
.0060	.0501	.0252	.0539	.0700	.0032	.1497	.0845	-.0131
.0065	.0506	.0246	.0555	.0703	.0027	.1523	.0848	-.0134
.0071	.0511	.0241	.0571	.0707	.0024	.1550	.0851	-.0136
.0077	.0516	.0236	.0587	.0710	.0020	.1576	.0853	-.0139
.0083	.0521	.0230	.0603	.0713	.0016	.1603	.0855	-.0142
.0089	.0526	.0225	.0620	.0717	.0012	.1630	.0858	-.0145
.0095	.0531	.0220	.0637	.0720	.0008	.1658	.0860	-.0148
.0102	.0536	.0214	.0654	.0723	.0004	.1685	.0863	-.0150
.0109	.0541	.0209	.0671	.0727	0	.1713	.0865	-.0153
.0116	.0546	.0204	.0689	.0730	-.0004	.1741	.0868	-.0156
.0123	.0550	.0198	.0707	.0733	-.0007	.1769	.0870	-.0158
.0131	.0555	.0193	.0725	.0736	-.0011	.1797	.0872	-.0161
.0139	.0560	.0188	.0743	.0740	-.0015	.1826	.0874	-.0163
.0147	.0565	.0183	.0761	.0743	-.0019	.1855	.0877	-.0166
.0155	.0569	.0178	.0780	.0747	-.0023	.1884	.0879	-.0168
.0163	.0574	.0172	.0799	.0750	-.0026	.1913	.0881	-.0171
.0172	.0578	.0167	.0818	.0754	-.0030	.1943	.0883	-.0173
.0181	.0583	.0162	.0837	.0757	-.0034	.1973	.0885	-.0175
.0190	.0587	.0157	.0857	.0760	-.0038	.2003	.0887	-.0178
.0200	.0591	.0152	.0877	.0764	-.0042	.2033	.0889	-.0180
.0209	.0596	.0147	.0897	.0767	-.0046	.2063	.0891	-.0182
.0219	.0600	.0142	.0917	.0771	-.0050	.2094	.0892	-.0184
.0229	.0604	.0137	.0938	.0779	-.0053	.2125	.0894	-.0187

[redacted]

TABLE I.- WING COORDINATES ALONG STREAMWISE CHORDS – Continued

(b) $\frac{y}{b/2} = 0.104$; c' = 45.839 cm (18.047 in.) – Concluded

x'/c'	z'/c' Upper surface	z'/c' Lower surface
0.2156	0.0896	-0.0189
.2187	.0897	-.0191
.2219	.0899	-.0193
.2251	.0900	-.0195
.2283	.0902	-.0197
.2315	.0903	-.0200
.2348	.0905	-.0201
.2380	.0906	-.0203
.2413	.0907	-.0205
.2446	.0908	-.0206
.2480	.0909	-.0208
.2513	.0910	-.0210
.2547	.0911	-.0212
.2581	.0911	-.0213
.2616	.0912	-.0215
.2650	.0912	-.0216
.2685	.0913	-.0218
.2720	.0913	-.0219
.2755	.0913	-.0220
.2791	.0913	-.0222
.2826	.0913	-.0223
.2862	.0913	-.0224
.2898	.0913	-.0225
.2935	.0912	-.0226
.2971	.0912	-.0227
.3008	.0912	-.0228
.3045	.0911	-.0229
.3082	.0910	-.0230
.3120	.0909	-.0231
.3157	.0908	-.0232
.3195	.0907	-.0232
.3233	.0906	
.3272	.0904	
.3310	.0903	
.3349	.0901	
.3388	.0900	
.3428	.0898	
.3467	.0897	
.3507	.0895	
.3547	.0893	-.0238
.3587	.0891	-.0238
.3627	.0888	-.0239
.3668	.0886	-.0239
.3709	.0884	-.0239
.3750	.0881	-.0239
.3791	.0879	-.0239
0.3833	0.0876	-0.0238
.3875	.0873	-.0238
.3917	.0871	-.0238
.3959	.0868	-.0237
.4001	.0865	-.0237
.4044	.0861	-.0236
.4087	.0858	-.0235
.4130	.0855	-.0234
.4173	.0851	-.0233
.4217	.0848	-.0232
.4261	.0844	-.0230
.4305	.0841	-.0230
.4817	.0793	-.0209
.5515	.0726	-.0155
.6095	.0669	-.0095
.6591	.0617	-.0035
.7023	.0571	.0018
.7406	.0530	.0065
.7748	.0493	.0105
.8057	.0454	.0135
.8341	.0414	.0157
.8602	.0374	.0168
.8843	.0336	.0170
.9068	.0299	.0163
.9277	.0265	.0153
.9474	.0232	.0140
.9659	.0196	.0124
.9747	.0178	.0115
.9833	.0159	.0103
.9943	.0132	.0088
.9966	.0126	.0085
1.0000		.0080

[redacted]

TABLE I.- WING COORDINATES ALONG STREAMWISE CHORDS – Continued

(c) $\frac{y}{b/2} = 0.158$; c' = 36.873 cm (14.517 in.)

x'/c'	z'/c' Upper surface	z'/c' Lower surface	x'/c'	z'/c' Upper surface	z'/c' Lower surface	x'/c'	z'/c' Upper surface	z'/c' Lower surface
0	0.0384	0.0384	0.1373	0.0834	-0.0104	0.2747	0.0881	-0.0191
.0030	.0472	.0294	.1403	.0837	-.0107	.2776	.0881	-.0191
.0060	.0507	.0257	.1433	.0839	-.0110	.2806	.0880	-.0192
.0090	.0533	.0229	.1463	.0842	-.0113	.2836	.0880	-.0193
.0119	.0554	.0206	.1493	.0844	-.0116	.2866	.0879	-.0194
.0149	.0572	.0186	.1523	.0846	-.0118	.2896	.0879	-.0194
.0179	.0588	.0168	.1552	.0848	-.0121	.2926	.0878	-.0195
.0209	.0602	.0152	.1582	.0851	-.0124	.2956	.0878	-.0196
.0239	.0615	.0137	.1612	.0853	-.0127	.2985	.0877	-.0196
.0269	.0626	.0124	.1642	.0855	-.0129	.3015	.0876	-.0197
.0299	.0636	.0112	.1672	.0857	-.0132	.3045	.0876	-.0197
.0328	.0646	.0100	.1702	.0859	-.0135	.3075	.0875	-.0198
.0358	.0656	.0090	.1732	.0860	-.0137	.3105	.0874	-.0198
.0388	.0665	.0079	.1761	.0862	-.0139	.3828	.0845	-.0192
.0418	.0674	.0070	.1791	.0864	-.0142	.4600	.0802	-.0158
.0448	.0683	.0061	.1821	.0865	-.0144	.5245	.0761	-.0115
.0478	.0691	.0053	.1851	.0867	-.0146	.5797	.0721	-.0068
.0508	.0698	.0045	.1881	.0868	-.0148	.6279	.0681	-.0021
.0537	.0706	.0037	.1911	.0869	-.0151	.6707	.0642	.0027
.0567	.0713	.0029	.1941	.0871	-.0153	.7095	.0605	.0071
.0597	.0719	.0022	.1970	.0872	-.0155	.7447	.0568	.0114
.0627	.0726	.0015	.2000	.0873	-.0157	.7769	.0533	.0153
.0657	.0732	.0008	.2030	.0874	-.0159	.8066	.0501	.0183
.0687	.0738	.0002	.2060	.0874	-.0161	.8341	.0470	.0205
.0717	.0744	-.0005	.2090	.0875	-.0162	.8596	.0438	.0219
.0746	.0750	-.0011	.2120	.0876	-.0164	.8836	.0402	.0225
.0776	.0756	-.0017	.2149	.0877	-.0166	.9075	.0368	.0225
.0806	.0761	-.0022	.2179	.0877	-.0167	.9313	.0333	.0220
.0836	.0766	-.0028	.2209	.0878	-.0169	.9548	.0295	.0207
.0866	.0771	-.0033	.2239	.0878	-.0170	.9663	.0276	.0199
.0896	.0775	-.0038	.2269	.0879	-.0172	.9776	.0255	.0190
.0925	.0780	-.0043	.2299	.0879	-.0174	.9924	.0226	.0174
.0955	.0784	-.0048	.2329	.0880	-.0175	.9956	.0220	.0170
.0985	.0789	-.0052	.2358	.0880	-.0176	1.0000		.0164
.1015	.0793	-.0057	.2388	.0881	-.0178			
.1045	.0797	-.0061	.2418	.0881	-.0179			
.1075	.0801	-.0066	.2448	.0882	-.0180			
.1105	.0804	-.0070	.2478	.0882	-.0182			
.1134	.0808	-.0074	.2508	.0882	-.0183			
.1164	.0811	-.0078	.2538	.0882	-.0184			
.1194	.0815	-.0082	.2567	.0882	-.0185			
.1224	.0818	-.0086	.2597	.0882	-.0186			
.1254	.0821	-.0090	.2627	.0882	-.0187			
.1284	.0825	-.0093	.2657	.0882	-.0188			
.1314	.0828	-.0097	.2687	.0882	-.0189			
.1343	.0831	-.0101	.2717	.0881	-.0190			

TABLE I.- WING COORDINATES ALONG STREAMWISE CHORDS – Continued

(d) $\frac{y}{b/2} = 0.232$; $c' = 26.355$ cm (10.376 in.)

x'/c'	z'/c'	
	Upper surface	Lower surface
0	0.0378	0.0378
.0006	.0420	.0336
.0028	.0471	.0284
.0055	.0507	.0246
.0096	.0545	.0206
.0150	.0581	.0167
.0204	.0607	.0136
.0335	.0654	.0080
.0648	.0723	0
.0942	.0764	-.0045
.1218	.0792	-.0073
.1727	.0826	-.0109
.2184	.0843	-.0130
.2987	.0860	-.0140
.3683	.0859	-.0128
.4293	.0847	-.0105
.4835	.0827	-.0077
.5322	.0805	-.0044
.5764	.0780	-.0007
.6174	.0751	.0033
.6575	.0725	.0081
.6962	.0699	.0129
.7335	.0669	.0178
.7694	.0641	.0228
.8043	.0611	.0273
.8382	.0579	.0310
.8713	.0544	.0334
.9038	.0506	.0347
.9361	.0461	.0346
.9522	.0438	.0339
.9683	.0414	.0329
.9893	.0381	.0308
.9937	.0374	.0302
1.0000		.0293

(e) $\frac{y}{b/2} = 0.309$; $c' = 20.808$ cm (8.192 in.)

x'/c'	z'/c'	
	Upper surface	Lower surface
0	0.0318	0.0318
.0004	.0349	.0288
.0018	.0385	.0251
.0035	.0411	.0224
.0062	.0439	.0196
.0097	.0465	.0169
.0131	.0486	.0147
.0217	.0529	.0107
.0426	.0597	.0046
.0626	.0642	.0007
.0819	.0677	-.0020
.1186	.0725	-.0063
.1530	.0758	-.0091
.2174	.0804	-.0117
.2770	.0828	-.0122
.3329	.0837	-.0117
.3857	.0838	-.0102
.4362	.0833	-.0081
.4846	.0822	-.0054
.5318	.0807	-.0020
.5781	.0794	.0023
.6236	.0780	.0076
.6682	.0763	.0139
.7120	.0745	.0209
.7549	.0722	.0283
.7972	.0696	.0353
.8388	.0665	.0408
.8799	.0628	.0441
.9204	.0582	.0449
.9404	.0554	.0441
.9604	.0523	.0424
.9866	.0475	.0385
.9921	.0464	.0374
1.0000		.0358

(f) $\frac{y}{b/2} = 0.386$; $c' = 18.654$ cm (7.344 in.)

x'/c'	z'/c'	
	Upper surface	Lower surface
0	0.0255	0.0255
.0002	.0282	.0229
.0013	.0314	.0196
.0025	.0337	.0173
.0044	.0361	.0148
.0069	.0384	.0124
.0094	.0402	.0106
.0156	.0435	.0074
.0310	.0492	.0024
.0462	.0532	-.0010
.0613	.0563	-.0037
.0911	.0610	-.0076
.1204	.0646	-.0103
.1777	.0697	-.0129
.2334	.0728	-.0141
.2876	.0750	-.0139
.3405	.0765	-.0129
.3922	.0773	-.0111
.4430	.0775	-.0088
.4927	.0775	-.0057
.5417	.0773	-.0017
.5898	.0768	.0034
.6373	.0760	.0099
.6842	.0749	.0179
.7306	.0733	.0266
.7765	.0712	.0355
.8220	.0688	.0427
.8671	.0655	.0471
.9118	.0610	.0481
.9341	.0582	.0471
.9562	.0549	.0448
.9852	.0495	.0397
.9913	.0482	.0383
1.0000		.0364

TABLE I.- WING COORDINATES ALONG STREAMWISE CHORDS – Continued

(g) $\frac{y}{b/2} = 0.464$; $c' = 12.537$ cm (4.936 in.)

x'/c'	z'/c'	
	Upper surface	Lower surface
0	0.0193	0.0193
.0002	.0218	.0168
.0012	.0248	.0137
.0023	.0270	.0115
.0041	.0294	.0091
.0064	.0316	.0068
.0088	.0332	.0051
.0146	.0367	.0020
.0291	.0420	-.0029
.0435	.0459	-.0062
.0578	.0488	-.0087
.0862	.0530	-.0122
.1143	.0561	-.0145
.1697	.0610	-.0170
.2241	.0646	-.0177
.2776	.0673	-.0173
.3304	.0694	-.0161
.3824	.0707	-.0144
.4338	.0720	-.0119
.4846	.0729	-.0085
.5347	.0734	-.0045
.5841	.0735	.0008
.6329	.0732	.0075
.6810	.0725	.0156
.7284	.0714	.0250
.7752	.0700	.0341
.8213	.0677	.0416
.8669	.0646	.0464
.9118	.0603	.0476
.9341	.0576	.0464
.9562	.0542	.0438
.9852	.0487	.0388
.9913	.0474	.0374
1.0000		.0352

(h) $\frac{y}{b/2} = 0.541$; $c' = 16.231$ cm (6.390 in.)

x'/c'	z'/c'	
	Upper surface	Lower surface
0	0.0122	0.0122
.0002	.0146	.0097
.0011	.0175	.0068
.0023	.0197	.0046
.0040	.0219	.0023
.0063	.0240	.0001
.0086	.0257	-.0015
.0143	.0289	-.0046
.0286	.0343	-.0093
.0429	.0380	-.0123
.0570	.0410	-.0145
.0852	.0454	-.0177
.1132	.0488	-.0198
.1686	.0540	-.0215
.2232	.0578	-.0218
.2770	.0608	-.0211
.3300	.0633	-.0195
.3823	.0654	-.0173
.4338	.0672	-.0147
.4846	.0686	-.0112
.5347	.0694	-.0069
.5841	.0700	-.0013
.6329	.0704	.0055
.6810	.0702	.0141
.7284	.0696	.0235
.7752	.0683	.0326
.8213	.0663	.0402
.8669	.0635	.0452
.9118	.0592	.0463
.9341	.0565	.0452
.9562	.0531	.0427
.9852	.0475	.0376
.9913	.0462	.0362
1.0000		.0339

(i) $\frac{y}{b/2} = 0.580$; $c' = 15.624$ cm (6.151 in.)

x'/c'	z'/c'	
	Upper surface	Lower surface
0	0.0082	0.0082
.0002	.0106	.0058
.0011	.0135	.0029
.0023	.0156	.0008
.0040	.0178	-.0015
.0063	.0199	-.0037
.0086	.0216	-.0053
.0143	.0248	-.0082
.0286	.0301	-.0127
.0429	.0340	-.0157
.0570	.0369	-.0178
.0852	.0414	-.0207
.1132	.0450	-.0226
.1686	.0504	-.0241
.2232	.0545	-.0241
.2770	.0577	-.0230
.3300	.0603	-.0214
.3823	.0626	-.0191
.4338	.0646	-.0163
.4846	.0662	-.0126
.5347	.0673	-.0082
.5841	.0681	-.0025
.6329	.0687	.0045
.6810	.0688	.0129
.7284	.0686	.0225
.7752	.0676	.0318
.8213	.0656	.0395
.8669	.0628	.0444
.9118	.0586	.0455
.9341	.0559	.0445
.9562	.0525	.0421
.9782	.0484	.0385
.9913	.0454	.0354
1.0000		.0331

TABLE I.- WING COORDINATES ALONG STREAMWISE CHORDS – Continued

(j) $\frac{y}{b/2} = 0.618$;

c' = 15.019 cm (5.913 in.)

x'/c'	z'/c' Upper surface	z'/c' Lower surface
0	0.0043	0.0043
.0002	.0066	.0019
.0012	.0095	-.0010
.0023	.0116	-.0031
.0040	.0138	-.0053
.0063	.0158	-.0075
.0086	.0174	-.0091
.0143	.0205	-.0119
.0286	.0259	-.0164
.0429	.0298	-.0194
.0570	.0328	-.0213
.0852	.0374	-.0239
.1132	.0411	-.0255
.1686	.0467	-.0268
.2232	.0509	-.0265
.2770	.0543	-.0251
.3300	.0571	-.0233
.3823	.0597	-.0209
.4338	.0618	-.0180
.4846	.0637	-.0142
.5347	.0650	-.0097
.5841	.0661	-.0039
.6329	.0669	.0033
.6810	.0673	.0117
.7284	.0672	.0216
.7752	.0664	.0308
.8213	.0647	.0385
.8669	.0620	.0437
.9118	.0579	.0446
.9341	.0551	.0436
.9562	.0518	.0414
.9782	.0476	.0376
.9913	.0445	.0346
1.0000		.0322

(k) $\frac{v}{b/2} = 0.657$;

c' = 14.412 cm (5.674 in.)

x'/c'	z'/c' Upper surface	z'/c' Lower surface
0	0.0002	0.0002
.0002	.0025	-.0022
.0011	.0054	-.0050
.0023	.0074	-.0071
.0040	.0096	-.0093
.0063	.0116	-.0114
.0086	.0131	-.0129
.0143	.0162	-.0160
.0286	.0218	-.0201
.0429	.0256	-.0230
.0570	.0287	-.0250
.0852	.0333	-.0273
.1132	.0370	-.0286
.1686	.0428	-.0294
.2232	.0472	-.0289
.2770	.0507	-.0274
.3300	.0538	-.0253
.3823	.0564	-.0229
.4338	.0588	-.0199
.4846	.0608	-.0159
.5347	.0626	-.0112
.5841	.0639	-.0053
.6329	.0650	.0018
.6810	.0656	.0106
.7284	.0657	.0205
.7752	.0650	.0297
.8213	.0636	.0374
.8669	.0610	.0425
.9118	.0570	.0436
.9341	.0543	.0426
.9562	.0509	.0404
.9782	.0467	.0367
.9913	.0436	.0336
1.0000		.0312

(l) $\frac{y}{b/2} = 0.696$;

c' = 13.807 cm (5.436 in.)

x'/c'	z'/c' Upper surface	z'/c' Lower surface
0	-0.0041	-0.0041
.0002	-.0018	-.0064
.0011	.0010	-.0092
.0023	.0031	-.0112
.0040	.0052	-.0134
.0063	.0072	-.0155
.0086	.0087	-.0170
.0143	.0119	-.0201
.0286	.0175	-.0241
.0429	.0214	-.0266
.0570	.0244	-.0285
.0852	.0291	-.0306
.1132	.0328	-.0318
.1686	.0388	-.0325
.2232	.0434	-.0317
.2770	.0469	-.0299
.3300	.0503	-.0277
.3823	.0530	-.0251
.4338	.0557	-.0219
.4846	.0579	-.0178
.5347	.0600	-.0129
.5841	.0616	-.0069
.6329	.0627	.0004
.6810	.0636	.0093
.7284	.0640	.0191
.7752	.0636	.0284
.8213	.0623	.0360
.8669	.0599	.0411
.9118	.0559	.0424
.9341	.0532	.0414
.9562	.0498	.0393
.9782	.0457	.0357
.9913	.0426	.0326
1.0000		.0302

TABLE I.- WING COORDINATES ALONG STREAMWISE CHORDS – Continued

(m) $\frac{y}{b/2} = 0.734$;

c' = 13.200 cm (5.197 in.)

x'/c'	z'/c' Upper surface	z'/c' Lower surface
0	-0.0085	-0.0085
.0002	-.0063	-.0108
.0011	-.0035	-.0136
.0023	-.0015	-.0156
.0040	.0006	-.0177
.0063	.0025	-.0197
.0086	.0041	-.0212
.0143	.0074	-.0239
.0286	.0130	-.0282
.0429	.0170	-.0305
.0570	.0199	-.0321
.0852	.0246	-.0340
.1132	.0285	-.0352
.1686	.0347	-.0357
.2232	.0394	-.0347
.2770	.0432	-.0327
.3300	.0467	-.0304
.3823	.0497	-.0274
.4338	.0525	-.0240
.4846	.0550	-.0198
.5347	.0572	-.0148
.5841	.0590	-.0086
.6329	.0603	-.0012
.6810	.0614	.0077
.7284	.0620	.0174
.7752	.0619	.0268
.8213	.0608	.0345
.8669	.0587	.0396
.9118	.0546	.0409
.9341	.0520	.0400
.9562	.0485	.0379
.9782	.0445	.0344
.9913	.0414	.0314
1.0000		.0290

(n) $\frac{y}{b/2} = 0.773$;

c' = 12.596 cm (4.959 in.)

x'/c'	z'/c' Upper surface	z'/c' Lower surface
0	-0.0128	-0.0128
.0002	-.0106	-.0151
.0011	-.0078	-.0178
.0023	-.0059	-.0198
.0040	-.0038	-.0219
.0063	-.0019	-.0239
.0086	-.0002	-.0253
.0143	.0029	-.0282
.0286	.0083	-.0322
.0429	.0121	-.0344
.0570	.0151	-.0361
.0852	.0199	-.0377
.1132	.0241	-.0387
.1686	.0305	-.0389
.2232	.0353	-.0377
.2770	.0393	-.0356
.3306	.0430	-.0332
.3823	.0461	-.0300
.4338	.0491	-.0263
.4846	.0518	-.0220
.5347	.0541	-.0169
.5841	.0561	-.0105
.6329	.0578	-.0031
.6810	.0590	.0058
.7284	.0596	.0154
.7752	.0598	.0249
.8213	.0590	.0328
.8669	.0569	.0378
.9118	.0531	.0393
.9341	.0505	.0384
.9562	.0471	.0364
.9782	.0431	.0330
.9913	.0398	.0301
1.0000		.0276

(o) $\frac{y}{b/2} = 0.812$;

c' = 11.989 cm (4.720 in.)

x'/c'	z'/c' Upper surface	z'/c' Lower surface
0	-0.0173	-0.0173
.0002	-.0151	-.0196
.0011	-.0124	-.0223
.0023	-.0105	-.0242
.0040	-.0085	-.0263
.0063	-.0066	-.0282
.0086	-.0049	-.0296
.0143	-.0018	-.0325
.0286	.0035	-.0363
.0429	.0073	-.0384
.0570	.0104	-.0400
.0852	.0153	-.0416
.1132	.0194	-.0425
.1686	.0258	-.0425
.2232	.0309	-.0411
.2770	.0352	-.0388
.3300	.0389	-.0362
.3823	.0422	-.0329
.4338	.0454	-.0288
.4846	.0482	-.0244
.5347	.0506	-.0191
.5841	.0528	-.0127
.6329	.0548	-.0052
.6810	.0561	.0037
.7284	.0570	.0133
.7752	.0574	.0227
.8213	.0568	.0306
.8669	.0548	.0357
.9118	.0514	.0374
.9341	.0489	.0366
.9562	.0455	.0347
.9782	.0414	.0314
.9913	.0382	.0286
1.0000		.0261

[redacted]

TABLE I.- WING COORDINATES ALONG STREAMWISE CHORDS – Concluded

(p) $\dfrac{y}{b/2} = 0.850$;
c' = 11.382 cm (4.481 in.)

x'/c'	z'/c'	
	Upper surface	Lower surface
0	-0.0221	-0.0221
.0002	-.0199	-.0243
.0011	-.0172	-.0270
.0023	-.0153	-.0289
.0040	-.0133	-.0309
.0063	-.0115	-.0328
.0086	-.0099	-.0342
.0143	-.0068	-.0368
.0286	-.0014	-.0405
.0429	.0025	-.0425
.0570	.0057	-.0440
.0852	.0105	-.0458
.1132	.0144	-.0467
.1686	.0208	-.0465
.2232	.0261	-.0448
.2770	.0304	-.0424
.3300	.0344	-.0394
.3823	.0379	-.0361
.4338	.0411	-.0318
.4846	.0440	-.0271
.5347	.0466	-.0217
.5841	.0493	-.0153
.6329	.0513	-.0077
.6810	.0529	.0014
.7284	.0541	.0109
.7752	.0547	.0202
.8213	.0542	.0282
.8669	.0526	.0334
.9118	.0494	.0353
.9341	.0469	.0347
.9562	.0438	.0329
.9782	.0397	.0296
.9913	.0367	.0269
1.0000		.0246

(q) $\dfrac{y}{b/2} = 0.935$;
c' = 10.051 cm (3.957 in.)

x'/c'	z'/c'	
	Upper surface	Lower surface
0	-0.0330	-0.0330
.0002	-.0309	-.0352
.0011	-.0283	-.0378
.0023	-.0264	-.0397
.0040	-.0245	-.0416
.0063	-.0227	-.0434
.0086	-.0213	-.0448
.0143	-.0185	-.0473
.0286	-.0133	-.0509
.0429	-.0094	-.0532
.0570	-.0064	-.0549
.0852	-.0014	-.0568
.1132	.0026	-.0574
.1686	.0090	-.0568
.2232	.0143	-.0546
.2770	.0189	-.0519
.3300	.0231	-.0481
.3823	.0271	-.0441
.4338	.0307	-.0395
.4846	.0340	-.0343
.5347	.0371	-.0287
.5841	.0401	-.0221
.6329	.0426	-.0141
.6810	.0447	-.0050
.7284	.0465	.0043
.7752	.0477	.0135
.8213	.0477	.0215
.8669	.0468	.0274
.9118	.0441	.0299
.9341	.0419	.0295
.9562	.0393	.0277
.9852	.0342	.0233
.9913	.0330	.0221
1.0000		.0202

(r) $\dfrac{y}{b/2} = 0.968$;
c' = 9.467 cm (3.727 in.)

x'/c'	z'/c'	
	Upper surface	Lower surface
0	-0.0382	-0.0382
.0002	-.0362	-.0405
.0011	-.0339	-.0432
.0023	-.0321	-.0450
.0040	-.0301	-.0467
.0063	-.0282	-.0484
.0086	-.0266	-.0497
.0143	-.0235	-.0522
.0286	-.0180	-.0558
.0429	-.0410	-.0580
.0570	-.0108	-.0598
.0852	-.0057	-.0618
.1132	-.0014	-.0622
.1686	.0052	-.0611
.2232	.0102	-.0590
.2770	.0147	-.0560
.3330	.0188	-.0521
.3823	.0230	-.0478
.4338	.0268	-.0428
.4846	.0303	-.0373
.5347	.0338	-.0314
.5841	.0368	-.0247
.6329	.0395	-.0166
.6810	.0417	-.0075
.7284	.0434	.0015
.7752	.0446	.0101
.8213	.0453	.0185
.8669	.0448	.0248
.9118	.0425	.0278
.9341	.0403	.0274
.9562	.0374	.0258
.9852	.0322	.0215
.9913	.0310	.0203
1.0000		.0184

[redacted]

TABLE II.- LOCATION OF PRESSURE ORIFICES ON MODEL

(a) Orifices on wing

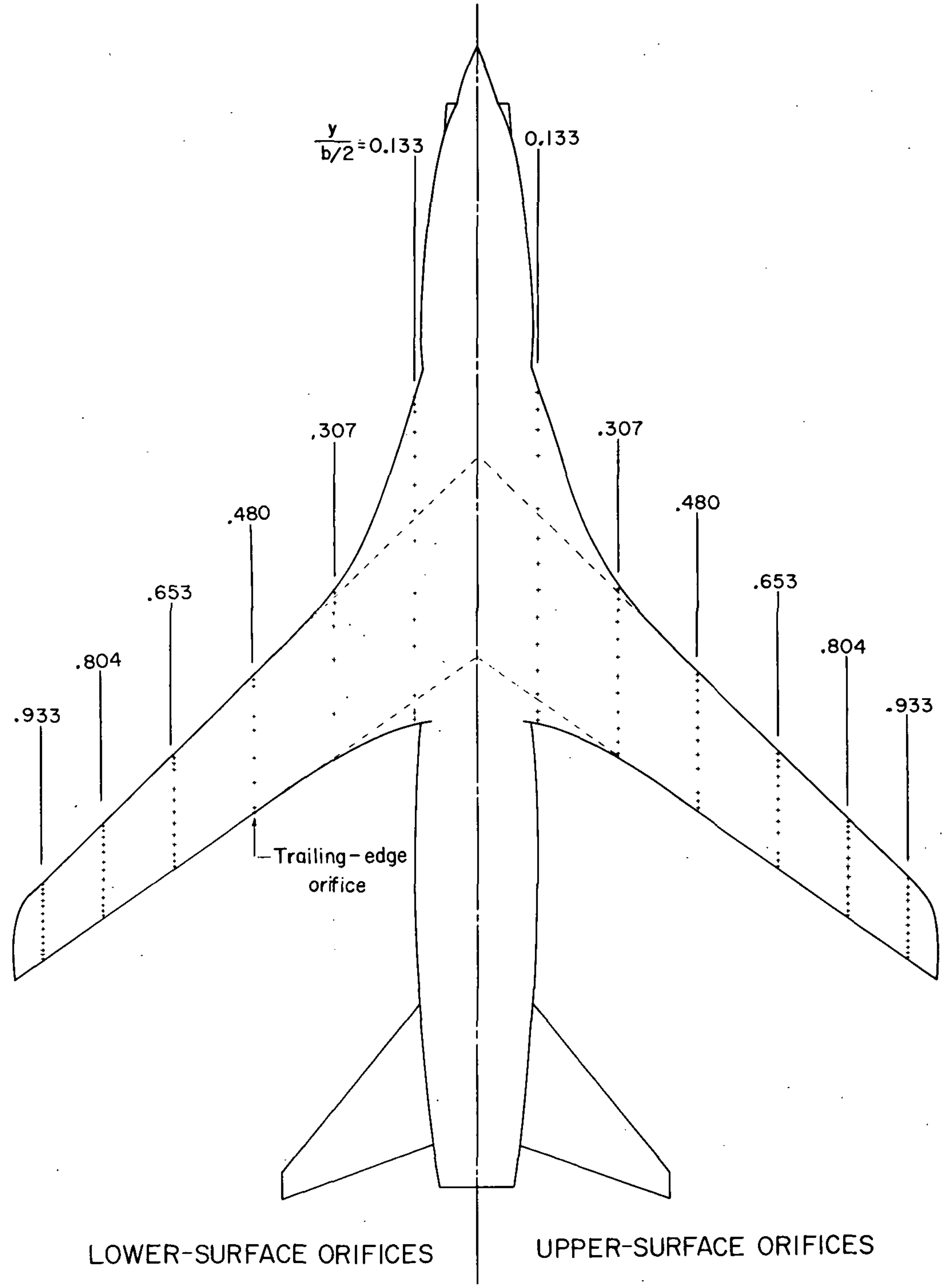

TABLE II.- LOCATION OF PRESSURE ORIFICES ON MODEL – Continued

(a) Orifices on wing – Concluded

[c in cm (in.)]

Wing orifice location, $\frac{x}{c}$, at semispan station, $\frac{y}{b/2}$, of –					
0.133 (c = 22.614 (8.903))	0.307 (c = 19.883 (7.828))	0.480 (c = 17.160 (6.756))	0.653 (c = 14.442 (5.686))	0.804 (c = 12.070 (4.752))	0.933 (c = 10.043 (3.954))
Right-wing upper surface					
-0.660	-0.021	0.023	0.025	0.022	0.018
-.567	.035	.068	.079	.075	.077
-.452	.105	.134	.133	.129	.129
-.311	.178	.209	.214	.201	.209
-.023	.286	.294	.295	.294	.293
.133	.396	.404	.407	.397	.494
.272	.514	.497	.502	.495	.590
.416	.618	.599	.601	.594	.693
.565	.733	.700	.698	.693	.777
.713	.835	.864	.863	.784	.861
.854	.919	.926	.923	.856	.918
.980	.987	.975	.977	.926	.972
1.074				.977	
1.122					
Left-wing lower surface					
-0.660	-0.022	0.024	0.025	0.019	0.020
-.616	.038	.075	.074	.066	.076
-.572	.101	.297	.130	.136	.136
-.462	.185	.400	.298	.214	.221
-.329	.398	.604	.397	.292	.295
-.172	.737	.785	.501	.403	.396
-.030		.967	.603	.489	.497
.128		1.000	.703	.594	.597
.418			.784	.700	.702
.564			.868	.786	.786
.710			.923	.858	.864
.976			.972	.919	.912
1.072				.967	.985
1.110					

[redacted]

TABLE II.- LOCATION OF PRESSURE ORIFICES ON MODEL – Continued

(b) Orifices on rear of fuselage

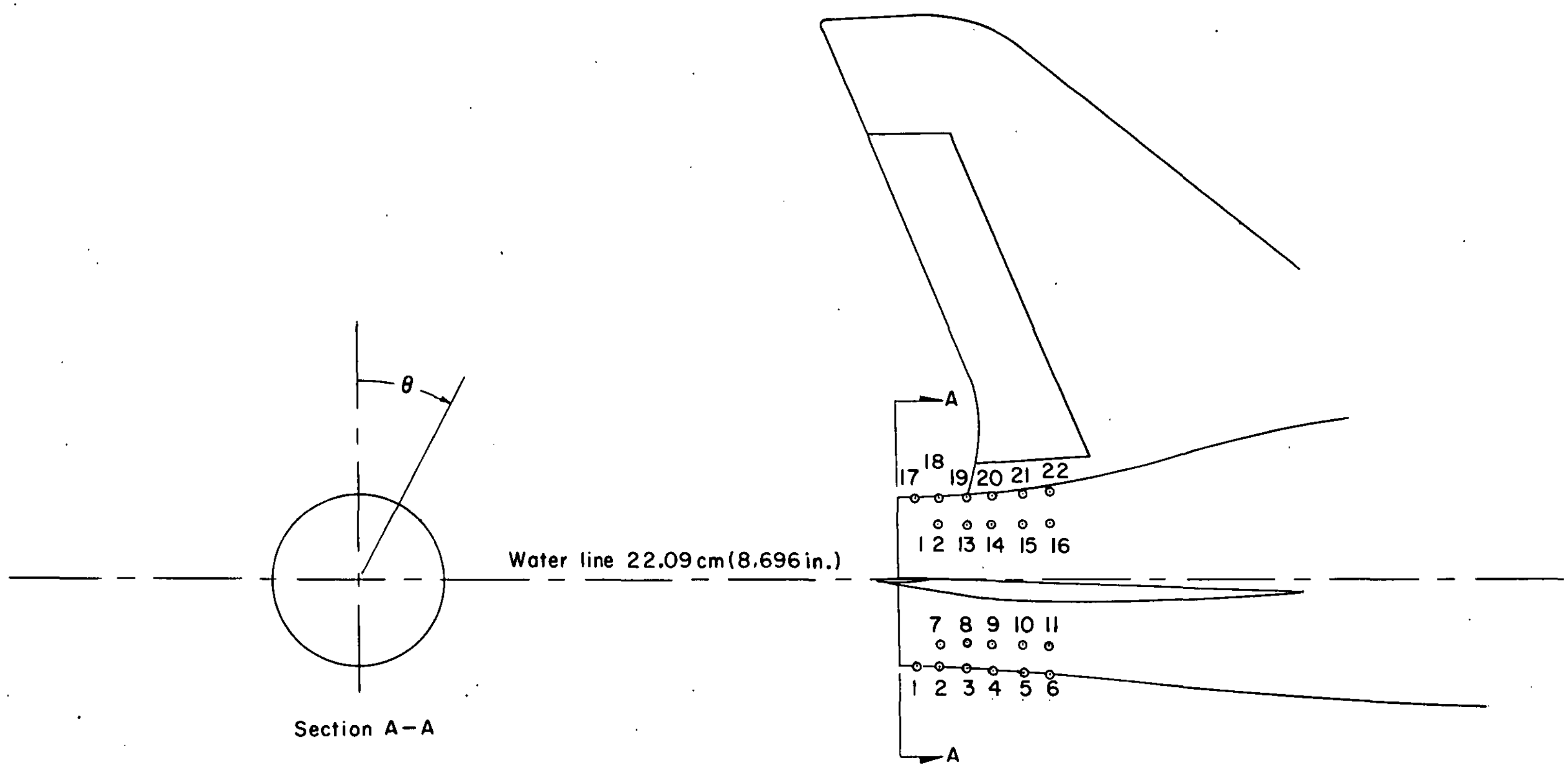

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TABLE II.- LOCATION OF PRESSURE ORIFICES ON MODEL – Continued

(b) Orifices on rear of fuselage – Concluded

Orifice	θ, deg	Model fuselage station	
		cm	in.
1	179.7	160.45	63.17
2	179.7	159.08	62.63
3	179.7	157.56	62.03
4	181.0	156.13	61.47
5	180.6	154.25	60.73
6	179.8	152.88	60.19
7	137.1	159.16	62.66
8	137.3	157.84	62.14
9	136.6	156.21	61.50
10	135.5	154.61	60.87
11	135.2	152.96	60.22
12	45.1	159.16	62.66
13	45.7	157.63	62.06
14	45.5	156.06	61.44
15	45.2	154.36	60.77
16	45.1	152.88	60.19
17	8.1	160.78	63.30
18	8.2	159.03	62.61
19	8.6	157.56	62.03
20	9.2	156.01	61.42
21	8.6	154.28	60.74
22	9.6	153.01	60.24

[redacted]

TABLE II.- LOCATION OF PRESSURE ORIFICES ON MODEL – Continued

(c) Orifices on area-rule additions

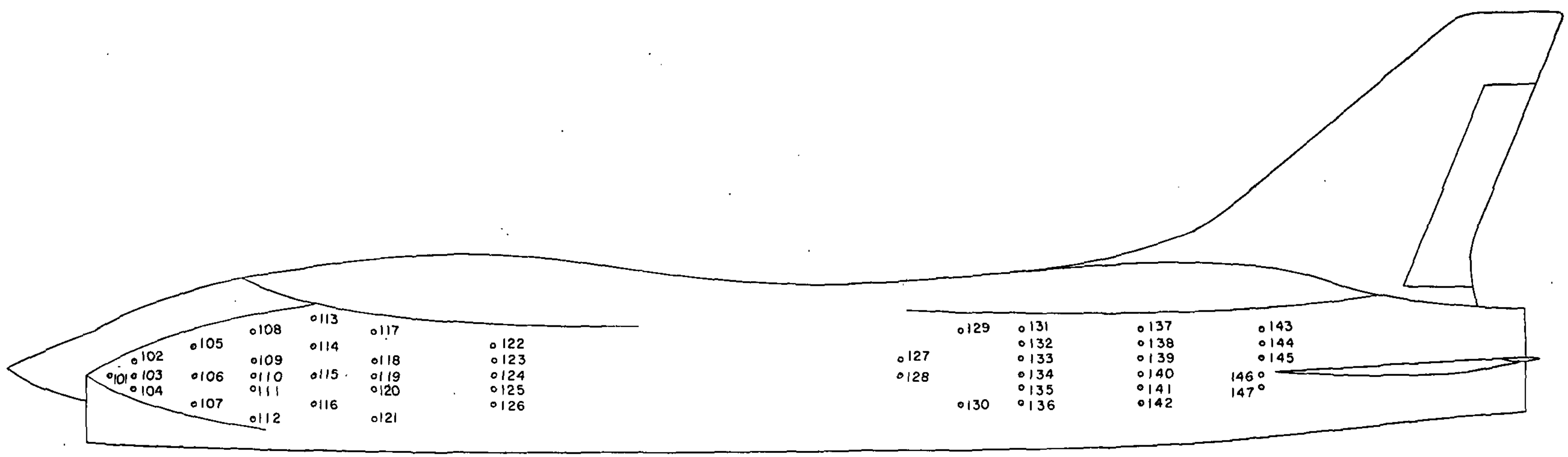

TABLE II.- LOCATION OF PRESSURE ORIFICES ON MODEL – Concluded

(c) Orifices on area-rule additions – Concluded

Fore fuselage area-rule addition					Aft fuselage area-rule addition				
Orifice	Model fuselage station		Model water line		Orifice	Model fuselage station		Model water line	
	cm	in.	cm	in.		cm	in.	cm	in.
101	30.92	12.17	20.75	8.17	127	103.81	40.87	22.10	8.70
102	33.12	13.04	22.10	8.70	128	103.81	40.87	20.75	8.17
103	33.12	13.04	20.75	8.17	129	109.32	43.04	24.74	9.74
104	33.12	13.04	19.43	7.65	130	109.32	43.04	18.11	7.13
105	38.66	15.22	23.42	9.22	131	114.88	45.22	24.74	9.74
106	38.66	15.22	20.75	8.17	132	114.86	45.22	23.42	9.22
107	38.66	15.22	18.11	7.13	133	114.86	45.22	22.10	8.70
108	44.17	17.39	24.74	9.74	134	114.86	45.22	20.75	8.17
109	44.17	17.39	22.10	8.70	135	114.86	45.22	19.43	7.65
110	44.17	17.39	20.75	8.17	136	114.86	45.22	18.11	7.13
111	44.17	17.39	19.43	7.65	137	125.91	49.57	24.74	9.74
112	44.17	17.39	16.79	6.61	138	125.91	49.57	23.42	9.22
113	49.71	19.57	26.06	10.26	139	125.91	49.57	22.10	8.70
114	49.71	19.57	23.42	9.22	140	125.91	49.57	20.75	8.17
115	49.71	19.57	20.75	8.17	141	125.91	49.57	19.43	7.65
116	49.71	19.57	18.11	7.13	142	125.91	49.57	18.11	7.13
117	55.22	21.74	24.74	9.74	143	136.93	53.91	24.74	9.74
118	55.22	21.74	22.10	8.70	144	136.93	53.91	23.42	9.22
119	55.22	21.74	20.75	8.17	145	136.93	53.91	22.10	8.70
120	55.22	21.74	19.43	7.65	146	136.93	53.91	20.75	8.17
121	55.22	21.74	16.79	6.61	147	136.93	53.91	19.43	7.65
122	66.27	26.09	23.42	9.22					
123	66.27	26.09	22.10	8.70					
124	66.27	26.09	20.75	8.17					
125	66.27	26.09	19.43	7.65					
126	66.27	26.09	18.11	7.13					

[redacted]

TABLE III. - TUNNEL TEST CONDITIONS

Mach number	Temperature		Reynolds number		Dynamic pressure	
	K	°F	per m	per ft	N/m^2	lb/ft^2
1.00	322	120	14.8	4.5	40 698	850
.99	322	120	14.8	4.5	40 698	850
.98	322	120	14.8	4.5	40 698	850
.97	322	120	14.8	4.5	40 698	850
.95	322	120	15.1	4.6	40 698	850
.90	322	120	15.7	4.8	40 698	850
.80	322	120	17.1	5.2	40 698	850
.50	322	120	13.1	4.0	21 546	450
.25	322	120	10.2	3.1	8 571	179

[redacted]

[redacted]

TABLE IV

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(a) M = 0.25.

$\alpha = -5.17^{\circ}$; $C_L = -0.432$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	.047	-.021	.340	.023	.432	.025	.439	.022	.501	.018	.525
	-.557	.007	.035	.206	.068	.281	.079	.309	.075	.348	.077	.321
	-.452	-.062	.105	.112	.134	.178	.133	.204	.129	.249	.129	.241
	-.311	-.089	.178	.056	.209	.112	.214	.149	.201	.191	.209	.161
	-.023	-.066	.286	.027	.294	.083	.295	.104	.294	.137	.293	.102
	.133	-.022	.396	-.001	.404	.029	.407	.053	.397	.090	.494	.016
	.272	.001	.514	-.007	.497	.001	.502	.027	.495	.041	.590	-.025
	.416	.020	.618	-.033	.599	-.026	.601	-.021	.594	-.006	.693	-.064
	.565	.035	.733	-.042	.700	-.064	.698	-.055	.693	-.055	.777	-.108
	.713	.023	.835	-.058	.864	-.115	.863	-.141	.784	-.102	.861	-.119
	.854	.001	.919	-.060	.926	-.110	.923	-.126	.856	-.139	.918	-.130
	.980	-.017	.987	-.042	.975	-.046	.977	-.059	.926	-.134	.972	-.060
	1.074	-.029							.977	-.043		
	1.122	-.024										
LOWER SURFACE												
	-.660	-.065	-.022	-1.184	.024	-1.749	.025	-1.944	.019	-2.801	.020	-1.771
	-.616	-.173	.038	-.780	.075	-1.009	.074	-1.030	.066	-1.205	.076	-1.485
	-.572	-.215	.101	-.673	.297	-.435	.130	-.755	.136	-.766	.136	-1.189
	-.462	-.234	.185	-.532	.400	-.342	.298	-.422	.214	-.540	.221	-.528
	-.329	-.243	.398	-.341	.604	-.203	.397	-.349	.292	-.449	.295	-.489
	-.172	-.273	.737	.000	.785	.055	.501	-.255	.403	-.343	.396	-.292
	-.030	-.306			.967	.165	.603	-.115	.489	-.275	.497	-.174
	.128	-.310			1.000	.045	.703	-.007	.594	-.198	.597	-.091
	.418	-.231					.784	.045	.700	-.046	.702	.013
	.564	-.164					.868	.126	.786	.013	.786	.035
	.710	-.044					.923	.147	.858	.070	.864	.045
	.976	.128					.972	.118	.919	.118	.912	.058
	1.110	.131							.967	.126	.985	.046
CN=		-.2577		-.3793		-.3904		-.3870		-.4638		-.4378
CM=		-.0923		-.0460		-.0631		-.0681		-.0674		-.0704

$\alpha = -4.17^{\circ}$; $C_L = -0.342$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	.048	-.021	.302	.023	.384	.025	.381	.022	.492	.018	.516
	-.557	-.013	.035	.144	.068	.198	.079	.249	.075	.282	.077	.281
	-.452	-.077	.105	.053	.134	.104	.133	.166	.129	.209	.129	.211
	-.311	-.114	.178	.016	.209	.063	.214	.113	.201	.140	.209	.135
	-.023	-.088	.286	-.010	.294	.033	.295	.062	.294	.090	.293	.070
	.133	-.040	.396	-.040	.404	.009	.407	.031	.397	.059	.494	.004
	.272	-.021	.514	-.029	.497	-.025	.502	-.006	.495	.007	.590	-.035
	.416	.003	.618	-.048	.599	-.053	.601	-.037	.594	-.016	.693	-.079
	.565	.013	.733	-.056	.700	-.084	.698	-.082	.693	-.070	.777	-.110
	.713	.012	.835	-.076	.864	-.129	.863	-.147	.784	-.107	.861	-.115
	.854	-.004	.919	-.069	.926	-.112	.923	-.125	.856	-.143	.918	-.122
	.980	-.017	.987	-.037	.975	-.045	.977	-.052	.926	-.134	.972	-.043
	1.074	-.022							.977	-.039		
	1.122	-.016										
LOWER SURFACE												
	-.660	-.053	-.022	-.878	.024	-1.367	.025	-1.675	.019	-2.222	.020	-2.300
	-.616	-.132	.038	-.704	.075	-.842	.074	-.884	.066	-1.044	.076	-.993
	-.572	-.181	.101	-.600	.297	-.382	.130	-.632	.136	-.676	.136	-.695
	-.462	-.193	.185	-.474	.400	-.314	.298	-.371	.214	-.491	.221	-.448
	-.329	-.205	.398	-.309	.604	-.188	.397	-.299	.292	-.392	.295	-.348
	-.172	-.236	.737	.019	.785	.059	.501	-.231	.403	-.295	.396	-.250
	-.030	-.282			.967	.174	.603	-.086	.489	-.243	.497	-.170
	.128	-.282			1.000	.049	.703	.021	.594	-.168	.597	-.073
	.418	-.207					.784	.064	.700	-.027	.702	.016
	.564	-.143					.868	.142	.786	.062	.786	.050
	.710	-.039					.923	.166	.858	.110	.864	.076
	.976	.139					.972	.135	.919	.137	.912	.071
	1.110	.130							.967	.125	.985	.063
CN=		-.1889		-.2966		-.2993		-.3012		-.3690		-.3548
CM=		-.0757		-.0464		-.0604		-.0708		-.0685		-.0717

[redacted]

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25. Continued.

$\alpha = -3.18^o$; $C_L = -0.250$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	.035	-.021	.270	.023	.263	.025	.277	.022	.456	.018	.496
	-.557	-.041	.035	.052	.068	.095	.079	.187	.075	.195	.077	.219
	-.452	-.106	.105	.001	.134	.067	.133	.089	.129	.152	.129	.133
	-.311	-.135	.178	-.050	.209	.004	.214	.063	.201	.096	.209	.092
	-.023	-.116	.286	-.077	.294	-.010	.295	.031	.294	.059	.293	.050
	.133	-.066	.396	-.061	.404	-.038	.407	-.005	.397	.031	.494	-.021
	.272	-.040	.514	-.066	.497	-.058	.502	-.033	.495	-.024	.590	-.057
	.416	-.009	.618	-.082	.599	-.089	.601	-.068	.594	-.047	.693	-.091
	.565	-.004	.733	-.076	.700	-.113	.698	-.096	.693	-.094	.777	-.119
	.713	.001	.835	-.088	.864	-.145	.863	-.158	.784	-.129	.861	-.121
	.854	-.033	.919	-.080	.926	-.121	.923	-.134	.856	-.158	.918	-.121
	.980	-.032	.987	-.046	.975	-.050	.977	-.045	.926	-.138	.972	-.041
	1.074	-.036							.977	-.037		
	1.122	-.020										
LOWER SURFACE												
	-.660	-.041	-.022	-.644	.024	-.999	.025	-1.239	.019	-1.544	.020	-1.845
	-.616	-.107	.038	-.602	.075	-.720	.074	-.704	.066	-.821	.076	-.793
	-.572	-.143	.101	-.489	.297	-.339	.130	-.531	.136	-.574	.136	-.595
	-.462	-.163	.185	-.406	.400	-.270	.298	-.316	.214	-.432	.221	-.409
	-.329	-.178	.398	-.273	.604	-.169	.397	-.264	.292	-.332	.295	-.296
	-.172	-.212	.737	.030	.785	.083	.501	-.209	.403	-.260	.396	-.224
	-.030	-.258			.967	.172	.603	-.086	.489	-.225	.497	-.169
	.128	-.266			1.000	.042	.703	.030	.594	-.154	.597	-.073
	.418	-.184					.784	.086	.700	-.001	.702	.039
	.554	-.125					.868	.152	.786	.091	.786	.111
	.710	-.023					.923	.177	.858	.145	.864	.120
	.976	.145					.972	.135	.919	.166	.912	.132
	1.110	.104							.967	.137	.985	.081
CN=	-.1187		-.2034		-.2024		-.2083		-.2623		-.2647	
CM=	-.0622		-.0469		-.0619		-.0682		-.0686		-.0743	

$\alpha = -2.21^o$; $C_L = -0.161$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	.021	-.021	.176	.023	.140	.025	.157	.022	.319	.018	.377
	-.567	-.055	.035	-.038	.068	.044	.079	.029	.075	.094	.077	.107
	-.452	-.124	.105	-.072	.134	-.022	.133	.006	.129	.059	.129	.073
	-.311	-.168	.178	-.096	.209	-.058	.214	-.027	.201	.023	.209	.025
	-.023	-.146	.286	-.105	.294	-.065	.295	-.043	.294	-.008	.293	-.008
	.133	-.084	.396	-.104	.404	-.076	.407	-.060	.397	-.044	.494	-.067
	.272	-.058	.514	-.094	.497	-.097	.502	-.086	.495	-.088	.590	-.094
	.416	-.045	.618	-.090	.599	-.105	.601	-.115	.594	-.096	.693	-.127
	.565	-.024	.733	-.094	.700	-.130	.698	-.149	.693	-.141	.777	-.152
	.713	-.016	.835	-.101	.864	-.162	.863	-.155	.784	-.168	.861	-.152
	.854	-.025	.919	-.094	.926	-.129	.923	-.164	.856	-.193	.918	-.149
	.980	-.044	.987	-.052	.975	-.062	.977	-.071	.926	-.168	.972	-.060
	1.074	-.041							.977	-.064		
	1.122	-.024										
LOWER SURFACE												
	-.660	-.010	-.022	-.432	.024	-.714	.025	-.968	.019	-1.225	.020	-1.040
	-.616	-.065	.038	-.467	.075	-.604	.074	-.629	.066	-.686	.076	-.730
	-.572	-.117	.101	-.405	.297	-.300	.130	-.460	.136	-.471	.136	-.475
	-.462	-.149	.185	-.380	.400	-.243	.298	-.283	.214	-.357	.221	-.333
	-.329	-.170	.398	-.242	.604	-.152	.397	-.228	.292	-.270	.295	-.256
	-.172	-.186	.737	.053	.785	.102	.501	-.185	.403	-.221	.396	-.205
	-.030	-.230			.967	.165	.603	-.078	.489	-.208	.497	-.146
	.128	-.239			1.000	.037	.703	.037	.594	-.139	.597	-.076
	.418	-.181					.784	.100	.700	.012	.702	.052
	.544	-.113					.868	.154	.786	.107	.786	.128
	.710	-.005					.923	.177	.858	.169	.864	.171
	.976	.152					.972	.137	.919	.188	.912	.160
	1.110	.101							.967	.140	.985	.084
CN=	-.0533		-.1188		-.1167		-.1072		-.1464		-.1411	
CM=	-.0492		-.0486		-.0621		-.0747		-.0764		-.0748	

[redacted]

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25. Continued.

$\alpha = -1.18^o$; $C_L = -0.064$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE											
-.660	.000	-.021	.064	.023	-.012	.025	.058	.022	.226	.018	.326
-.567	-.092	.035	-.128	.068	-.088	.079	-.045	.075	.010	.077	.035
-.452	-.169	.105	-.124	.134	-.132	.133	-.029	.129	.011	.129	.015
-.311	-.207	.178	-.146	.209	-.112	.214	-.059	.201	-.036	.209	-.012
-.023	-.174	.286	-.161	.294	-.110	.295	-.080	.294	-.059	.293	-.045
.133	-.121	.396	-.145	.404	-.114	.407	-.088	.397	-.057	.494	-.075
.272	-.085	.514	-.113	.497	-.124	.502	-.115	.495	-.076	.590	-.099
.416	-.068	.618	-.125	.599	-.129	.601	-.123	.594	-.102	.693	-.123
.565	-.036	.733	-.125	.700	-.161	.698	-.153	.693	-.145	.777	-.156
.713	-.028	.835	-.113	.864	-.178	.863	-.185	.784	-.176	.861	-.150
.854	-.045	.919	-.112	.926	-.143	.923	-.156	.856	-.185	.918	-.146
.980	-.050	.987	-.052	.975	-.064	.977	-.061	.926	-.164	.972	-.055
1.074	-.049							.977	-.053		
1.122	-.029										
LOWER SURFACE											
-.660	.021	-.022	-.188	.024	-.469	.025	-.699	.019	-.829	.020	-.754
-.616	-.061	.038	-.310	.075	-.443	.074	-.511	.066	-.635	.076	-.577
-.572	-.076	.101	-.350	.297	-.242	.130	-.372	.136	-.372	.136	-.456
-.462	-.123	.185	-.293	.400	-.196	.298	-.231	.214	-.274	.221	-.249
-.329	-.136	.398	-.196	.604	-.131	.397	-.196	.292	-.228	.295	-.204
-.172	-.171	.737	.068	.785	.110	.501	-.158	.403	-.196	.396	-.179
-.030	-.210			.967	.168	.603	-.064	.489	-.170	.497	-.131
.128	-.213			1.000	.034	.703	.040	.594	-.121	.597	-.065
.418	-.148					.784	.123	.700	.020	.702	.061
.564	-.090					.868	.170	.786	.124	.786	.138
.710	.013					.923	.194	.858	.191	.864	.191
.976	.152					.972	.143	.919	.209	.912	.189
1.110	.105							.967	.142	.985	.076
CN=	.0299		-.0150		-.0145		-.0343		-.0736		-.0775
CM=	-.0326		-.0493		-.0615		-.0720		-.0733		-.0724

$\alpha = -0.05^o$; $C_L = 0.041$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE											
-.660	-.029	-.021	-.096	.023	-.138	.025	-.155	.022	.019	.018	.125
-.567	-.134	.035	-.210	.068	-.232	.079	-.153	.075	-.092	.077	-.071
-.452	-.215	.105	-.215	.134	-.179	.133	-.138	.129	-.104	.129	-.055
-.311	-.239	.178	-.212	.209	-.166	.214	-.142	.201	-.100	.209	-.084
-.023	-.198	.296	-.198	.294	-.153	.295	-.142	.294	-.104	.293	-.079
.133	-.139	.396	-.179	.404	-.159	.407	-.135	.397	-.099	.494	-.104
.272	-.103	.514	-.149	.497	-.161	.502	-.145	.495	-.129	.590	-.130
.416	-.081	.618	-.158	.599	-.162	.601	-.174	.594	-.149	.693	-.153
.565	-.064	.733	-.135	.700	-.186	.698	-.189	.693	-.178	.777	-.165
.713	-.049	.835	-.130	.864	-.193	.863	-.216	.784	-.204	.861	-.173
.854	-.062	.919	-.113	.926	-.149	.923	-.170	.856	-.212	.918	-.159
.980	-.054	.987	-.049	.975	-.069	.977	-.061	.926	-.182	.972	-.064
1.074	-.061							.977	-.068		
1.122	-.026										
LOWER SURFACE											
-.660	.030	-.022	-.031	.024	-.245	.025	-.423	.019	-.503	.020	-.505
-.616	-.030	.038	-.235	.075	-.313	.074	-.333	.066	-.396	.076	-.410
-.572	-.045	.101	-.233	.297	-.214	.130	-.280	.136	-.301	.136	-.318
-.462	-.105	.185	-.231	.400	-.153	.298	-.176	.214	-.230	.221	-.237
-.329	-.120	.398	-.155	.604	-.114	.397	-.151	.292	-.182	.295	-.152
-.172	-.159	.737	.070	.785	.127	.501	-.129	.403	-.142	.396	-.126
-.030	-.184			.967	.166	.603	-.047	.489	-.137	.497	-.107
.128	-.190			1.000	.025	.703	.043	.594	-.096	.597	-.051
.418	-.124					.784	.131	.700	.043	.702	.068
.564	-.075					.868	.193	.786	.143	.786	.144
.710	.025					.923	.222	.858	.210	.864	.212
.976	.163					.972	.155	.919	.231	.912	.212
1.110	.103							.967	.143	.985	.065
CN=	.1045		.0742		.0740		.0790		.0403		.0167
CM=	-.0193		-.0463		-.0625		-.0750		-.0772		-.0725

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25. Continued.

$\alpha = 0.93^o$; $C_L = 0.127$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.650	-.049	-.021	-.249	.023	-.364	.025	-.242	.022	-.114	.018	-.005
-.567	-.189	.035	-.317	.068	-.310	.079	-.269	.075	-.197	.077	-.166
-.452	-.235	.105	-.317	.134	-.257	.133	-.219	.129	-.184	.129	-.126
-.311	-.259	.178	-.281	.209	-.238	.214	-.192	.201	-.146	.209	-.117
-.023	-.221	.286	-.253	.294	-.217	.295	-.174	.294	-.152	.293	-.121
.133	-.158	.396	-.219	.404	-.187	.407	-.168	.397	-.152	.494	-.133
.272	-.136	.514	-.175	.497	-.197	.502	-.174	.495	-.166	.590	-.148
.416	-.098	.618	-.172	.599	-.193	.601	-.193	.594	-.172	.693	-.172
.565	-.069	.733	-.146	.700	-.195	.698	-.207	.693	-.201	.777	-.188
.713	-.061	.835	-.141	.864	-.197	.863	-.224	.784	-.220	.861	-.181
.854	-.062	.919	-.101	.926	-.152	.923	-.174	.856	-.219	.918	-.168
.980	-.061	.987	-.050	.975	-.065	.977	-.059	.926	-.191	.972	-.076
1.074	-.058							.977	-.064		
1.122	-.038										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.650	.045	-.022	.081	.024	-.052	.025	-.153	.019	-.240	.020	-.212
-.616	-.002	.038	-.115	.075	-.191	.074	-.197	.066	-.255	.076	-.306
-.572	-.010	.101	-.166	.297	-.153	.130	-.187	.136	-.211	.136	-.256
-.462	-.069	.185	-.159	.400	-.125	.298	-.141	.214	-.160	.221	-.189
-.329	-.093	.398	-.135	.604	-.091	.397	-.108	.292	-.142	.295	-.155
-.172	-.110	.737	.082	.785	.144	.501	-.102	.403	-.102	.396	-.106
-.030	-.153			.967	.168	.603	-.037	.489	-.108	.497	-.085
.128	-.153			1.000	.026	.703	.045	.594	-.082	.597	-.042
.418	-.098					.784	.127	.700	.050	.702	.074
.564	-.039					.868	.211	.786	.152	.786	.153
.710	.047					.923	.250	.858	.234	.864	.226
.976	.178					.972	.165	.919	.245	.912	.227
1.110	.113							.967	.150	.985	.065

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.1884	.1603	.1643	.1607	.1290	.0892
CM=	-.0024	-.0429	-.0615	-.0743	-.0781	-.0723

$\alpha = 2.03^o$; $C_L = 0.223$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.650	-.101	-.021	-.414	.023	-.630	.025	-.471	.022	-.333	.018	-.206
-.567	-.220	.035	-.474	.068	-.485	.079	-.382	.075	-.309	.077	-.286
-.452	-.287	.105	-.377	.134	-.351	.133	-.305	.129	-.255	.129	-.202
-.311	-.286	.178	-.347	.209	-.301	.214	-.253	.201	-.235	.209	-.181
-.023	-.252	.286	-.311	.294	-.262	.295	-.218	.294	-.202	.293	-.161
.133	-.184	.396	-.257	.404	-.236	.407	-.196	.397	-.177	.494	-.150
.272	-.154	.514	-.210	.497	-.230	.502	-.202	.495	-.187	.590	-.167
.416	-.124	.618	-.190	.599	-.210	.601	-.189	.594	-.191	.693	-.181
.565	-.092	.733	-.173	.700	-.221	.698	-.226	.693	-.208	.777	-.208
.713	-.074	.835	-.152	.864	-.222	.863	-.219	.784	-.227	.861	-.181
.854	-.088	.919	-.118	.926	-.156	.923	-.179	.856	-.230	.918	-.164
.980	-.085	.987	-.050	.975	-.070	.977	-.060	.926	-.183	.972	-.063
1.074	-.072							.977	-.062		
1.122	-.044										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.650	.042	-.022	.219	.024	.092	.025	.057	.019	.007	.020	.014
-.616	.018	.038	.002	.075	-.070	.074	-.095	.066	-.142	.076	-.186
-.572	-.006	.101	-.087	.297	-.110	.130	-.094	.136	-.124	.136	-.176
-.462	-.039	.185	-.114	.400	-.093	.298	-.109	.214	-.109	.221	-.135
-.329	-.069	.398	-.094	.604	-.074	.397	-.073	.292	-.092	.295	-.127
-.172	-.092	.737	.090	.785	.150	.501	-.081	.403	-.073	.396	-.076
-.030	-.124			.967	.165	.603	-.014	.489	-.087	.497	-.072
.128	-.147			1.000	.020	.703	.050	.594	-.060	.597	-.039
.418	-.080					.784	.137	.700	.058	.702	.071
.564	-.027					.868	.213	.786	.155	.786	.155
.710	.058					.923	.250	.858	.231	.864	.215
.976	.178					.972	.163	.919	.245	.912	.226
1.110	.108							.967	.139	.985	.057

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.2592	.2536	.2590	.2409	.2102	.1614
CM=	.0110	-.0417	-.0604	-.0701	-.0735	-.0660

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25. Continued.

$\alpha = 2.48^o$; $C_L = 0.267$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.109	-.021	-.481	.023	-.667	.025	-.568	.022	-.486	.018	-.288
-.567	-.238	.035	-.485	.068	-.539	.079	-.431	.075	-.378	.077	-.345
-.452	-.271	.105	-.424	.134	-.395	.133	-.338	.129	-.307	.129	-.255
-.311	-.317	.178	-.380	.209	-.333	.214	-.291	.201	-.253	.209	-.204
-.023	-.260	.286	-.329	.294	-.288	.295	-.252	.294	-.221	.293	-.185
.133	-.186	.396	-.259	.404	-.246	.407	-.229	.397	-.201	.494	-.161
.272	-.166	.514	-.202	.497	-.238	.502	-.221	.495	-.212	.590	-.176
.416	-.118	.618	-.190	.599	-.228	.601	-.221	.594	-.211	.693	-.190
.565	-.092	.733	-.166	.700	-.228	.698	-.221	.693	-.225	.777	-.208
.713	-.078	.835	-.154	.864	-.207	.863	-.241	.784	-.229	.861	-.190
.854	-.084	.919	-.110	.926	-.158	.923	-.181	.856	-.240	.918	-.176
.980	-.094	.987	-.040	.975	-.060	.977	-.067	.926	-.189	.972	-.071
1.074	-.070							.977	-.059		
1.122	-.038										
					LOWER SURFACE						
-.660	.051	-.022	.264	.024	.182	.025	.116	.019	.095	.020	.083
-.616	.034	.038	.028	.075	-.032	.074	-.033	.066	-.045	.076	-.104
-.572	.001	.101	-.045	.297	-.083	.130	-.061	.136	-.084	.136	-.155
-.462	-.023	.185	-.077	.400	-.065	.298	-.079	.214	-.062	.221	-.101
-.329	-.055	.398	-.081	.604	-.060	.397	-.058	.292	-.064	.295	-.101
-.172	-.086	.737	.096	.785	.157	.501	-.064	.403	-.053	.396	-.061
-.030	-.112			.967	.170	.603	.002	.489	-.070	.497	-.058
.128	-.117			1.000	.026	.703	.059	.594	-.056	.597	-.019
.418	-.069					.784	.148	.700	.069	.702	.082
.564	-.018					.868	.224	.786	.172	.786	.164
.710	.063					.923	.270	.858	.238	.864	.220
.976	.192					.972	.167	.919	.253	.912	.227
1.110	.116							.967	.145	.985	.050
CN=	.2946		.2839		.2973		.2905		.2638		.2036
CM=	.0176		-.0387		-.0602		-.0739		-.0744		-.0667

$\alpha = 2.88^o$; $C_L = 0.300$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.126	-.021	-.598	.023	-.758	.025	-.674	.022	-.523	.018	-.405
-.567	-.262	.035	-.537	.068	-.611	.079	-.468	.075	-.424	.077	-.395
-.452	-.309	.105	-.461	.134	-.409	.133	-.381	.129	-.350	.129	-.265
-.311	-.316	.178	-.386	.209	-.338	.214	-.321	.201	-.282	.209	-.229
-.023	-.258	.286	-.328	.294	-.301	.295	-.259	.294	-.249	.293	-.192
.133	-.196	.396	-.282	.404	-.261	.407	-.239	.397	-.206	.494	-.175
.272	-.162	.514	-.220	.497	-.244	.502	-.233	.495	-.212	.590	-.184
.416	-.141	.618	-.205	.599	-.232	.601	-.235	.594	-.226	.693	-.198
.565	-.120	.733	-.176	.700	-.233	.698	-.257	.693	-.234	.777	-.212
.713	-.086	.835	-.150	.864	-.208	.863	-.239	.784	-.245	.861	-.191
.854	-.086	.919	-.113	.926	-.162	.923	-.181	.856	-.242	.918	-.179
.980	-.085	.987	-.042	.975	-.069	.977	-.063	.926	-.187	.972	-.078
1.074	-.069							.977	-.058		
1.122	-.037										
					LOWER SURFACE						
-.660	.058	-.022	.265	.024	.258	.025	.188	.019	.149	.020	.143
-.616	.037	.038	.061	.075	.024	.074	.025	.066	-.048	.076	-.072
-.572	.038	.101	-.021	.297	-.067	.130	-.019	.136	-.049	.136	-.097
-.462	-.016	.185	-.035	.400	-.040	.298	-.056	.214	-.050	.221	-.097
-.329	-.044	.398	-.070	.604	-.051	.397	-.050	.292	-.062	.295	-.083
-.172	-.067	.737	.105	.785	.164	.501	-.056	.403	-.048	.396	-.049
-.030	-.098			.967	.169	.603	.008	.489	-.056	.497	-.058
.128	-.104			1.000	.020	.703	.059	.594	-.049	.597	-.022
.418	-.062					.784	.147	.700	.067	.702	.085
.564	-.009					.868	.228	.786	.163	.786	.155
.710	.080					.923	.272	.858	.239	.864	.214
.976	.197					.972	.166	.919	.251	.912	.226
1.110	.121							.967	.149	.985	.055
CN=	.3325		.3224		.3324		.3289		.2876		.2313
CM=	.0249		-.0393		-.0596		-.0736		-.0742		-.0648

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25. Continued.

$\alpha = 3.35^o$; $C_L = 0.341$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.151	-.021	-.731	.023	-.889	.025	-.761	.022	-.648	.018	-.499
-.567	-.291	.035	-.601	.068	-.699	.079	-.562	.075	-.503	.077	-.468
-.452	-.340	.105	-.501	.134	-.474	.133	-.421	.129	-.386	.129	-.313
-.311	-.331	.178	-.437	.209	-.401	.214	-.350	.201	-.319	.209	-.267
-.023	-.291	.286	-.351	.294	-.348	.295	-.292	.294	-.275	.293	-.231
.133	-.215	.396	-.311	.404	-.291	.407	-.267	.397	-.245	.494	-.189
.272	-.181	.514	-.240	.497	-.276	.502	-.255	.495	-.232	.590	-.206
.416	-.146	.618	-.220	.599	-.251	.601	-.255	.594	-.239	.693	-.212
.565	-.104	.733	-.185	.700	-.243	.698	-.266	.693	-.249	.777	-.216
.713	-.098	.835	-.159	.864	-.219	.863	-.248	.784	-.259	.861	-.201
.854	-.094	.919	-.115	.926	-.163	.923	-.185	.856	-.251	.918	-.189
.980	-.096	.987	-.050	.975	-.068	.977	-.067	.926	-.185	.972	-.078
1.074	-.090							.977	-.064		
1.122	-.049										
					LOWER SURFACE						
-.660	.043	-.022	.305	.024	.269	.025	.234	.019	.226	.020	.206
-.616	.041	.038	.097	.075	.060	.074	.044	.066	.034	.076	-.011
-.572	.031	.101	.011	.297	-.061	.130	.003	.136	-.017	.136	-.071
-.462	-.012	.185	-.035	.400	-.044	.298	-.046	.214	-.032	.221	-.078
-.329	-.028	.398	-.056	.604	-.041	.397	-.033	.292	-.031	.295	-.066
-.172	-.064	.737	.114	.785	.164	.501	-.044	.403	-.043	.396	-.046
-.030	-.100			.967	.165	.603	.015	.489	-.048	.497	-.048
.128	-.095			1.000	.016	.703	.060	.594	-.043	.597	-.026
.418	-.042					.784	.148	.700	.074	.702	.080
.564	-.011					.868	.227	.786	.179	.786	.157
.710	.081					.923	.273	.858	.242	.864	.220
.976	.198					.972	.170	.919	.255	.912	.217
1.110	.117							.967	.142	.985	.048
CN=	.3683		.3663		.3752		.3672		.3350		.2704
CM=	.0334		-.0404		-.0594		-.0746		-.0747		-.0638

$\alpha = 3.88^o$; $C_L = 0.388$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.183	-.021	-.875	.023	-1.013	.025	-.880	.022	-.875	.018	-.683
-.567	-.231	.035	-.653	.068	-.797	.079	-.646	.075	-.577	.077	-.524
-.452	-.344	.105	-.551	.134	-.505	.133	-.464	.129	-.437	.129	-.367
-.311	-.344	.178	-.453	.209	-.426	.214	-.377	.201	-.358	.209	-.277
-.023	-.292	.286	-.384	.294	-.361	.295	-.332	.294	-.302	.293	-.247
.133	-.230	.396	-.321	.404	-.304	.407	-.287	.397	-.255	.494	-.201
.272	-.182	.514	-.263	.497	-.283	.502	-.267	.495	-.251	.590	-.206
.416	-.151	.618	-.235	.599	-.248	.601	-.263	.594	-.243	.693	-.206
.565	-.118	.733	-.187	.700	-.251	.698	-.265	.693	-.256	.777	-.225
.713	-.105	.835	-.167	.864	-.215	.863	-.253	.784	-.260	.861	-.200
.854	-.107	.919	-.107	.926	-.155	.923	-.186	.856	-.255	.918	-.180
.980	-.101	.987	-.042	.975	-.064	.977	-.064	.926	-.193	.972	-.078
1.074	-.085							.977	-.057		
1.122	-.049										
					LOWER SURFACE						
-.660	.048	-.022	.342	.024	.343	.025	.284	.019	.286	.020	.269
-.616	.053	.038	.133	.075	.110	.074	.100	.066	.093	.076	.036
-.572	.042	.101	.044	.297	-.022	.130	.038	.136	.008	.136	-.046
-.462	.006	.185	-.009	.400	-.020	.298	-.016	.214	-.006	.221	-.042
-.329	-.017	.398	-.037	.604	-.034	.397	-.022	.292	-.027	.295	-.047
-.172	-.049	.737	.112	.785	.171	.501	-.028	.403	-.005	.396	-.036
-.030	-.070			.967	.166	.603	.024	.489	-.029	.497	-.041
.128	-.082			1.000	.006	.703	.077	.594	-.020	.597	-.007
.418	-.029					.784	.154	.700	.077	.702	.084
.564	.004					.868	.232	.786	.170	.786	.154
.710	.086					.923	.274	.858	.240	.864	.217
.976	.205					.972	.170	.919	.257	.912	.219
1.110	.123							.967	.146	.985	.045
CN=	.4047		.4064		.4167		.4108		.3812		.3061
CM=	.0366		-.0372		-.0571		-.0741		-.0726		-.0606

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(a) M = 0.25. Continued.

$\alpha = 4.89^{\circ}$; $C_L = 0.474$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.204	-.021	-1.090	.023	-1.306	.025	-1.089	.022	-1.100	.018	-.906
-.567	-.349	.035	-.802	.068	-.775	.079	-.774	.075	-.700	.077	-.648
-.452	-.377	.105	-.621	.134	-.617	.133	-.562	.129	-.511	.129	-.448
-.311	-.388	.178	-.526	.209	-.492	.214	-.448	.201	-.428	.209	-.347
-.023	-.311	.286	-.422	.294	-.394	.295	-.374	.294	-.351	.293	-.288
.133	-.252	.396	-.357	.404	-.328	.407	-.322	.397	-.298	.494	-.216
.272	-.200	.514	-.272	.497	-.299	.502	-.302	.495	-.279	.590	-.230
.416	-.170	.618	-.247	.599	-.280	.601	-.288	.594	-.277	.693	-.234
.565	-.127	.733	-.191	.700	-.259	.698	-.281	.693	-.272	.777	-.234
.713	-.121	.835	-.158	.864	-.212	.863	-.257	.784	-.273	.861	-.216
.854	-.121	.919	-.102	.926	-.150	.923	-.189	.856	-.256	.918	-.185
.930	-.109	.987	-.040	.975	-.064	.977	-.059	.926	-.190	.972	-.083
1.074	-.089							.977	-.057		
1.122	-.054										
				LOWER SURFACE							
-.660	.043	-.022	.374	.024	.432	.025	.390	.019	.395	.020	.389
-.616	.073	.038	.210	.075	.190	.074	.166	.066	.154	.076	.131
-.572	.050	.101	.108	.297	.026	.130	.112	.136	.095	.136	.040
-.462	.029	.185	.053	.400	.008	.298	.020	.214	.057	.221	-.009
-.329	.012	.398	-.005	.604	-.014	.397	.027	.292	.032	.295	-.012
-.172	-.012	.737	.129	.785	.182	.501	-.004	.403	.009	.396	-.014
-.030	-.042			.967	.169	.603	.043	.489	-.003	.497	-.028
.128	-.050			1.000	.007	.703	.086	.594	-.009	.597	-.003
.418	-.005					.784	.162	.700	.091	.702	.080
.564	.029					.868	.233	.786	.179	.786	.154
.710	.102					.923	.278	.858	.253	.864	.215
.976	.223					.972	.166	.919	.260	.912	.217
1.110	.130							.967	.144	.985	.039
CN=	.4883		.4869		.4866		.4897		.4601		.3792
CM=	.0528		-.0342		-.0563		-.0734		-.0724		-.0569

$\alpha = 5.96^{\circ}$; $C_L = 0.564$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.289	-.021	-1.361	.023	-1.697	.025	-1.506	.022	-1.482	.018	-1.262
-.567	-.387	.035	-.922	.068	-1.002	.079	-.957	.075	-.871	.077	-.776
-.452	-.411	.105	-.711	.134	-.686	.133	-.638	.129	-.644	.129	-.547
-.311	-.424	.178	-.602	.209	-.562	.214	-.513	.201	-.492	.209	-.403
-.023	-.345	.286	-.477	.294	-.471	.295	-.429	.294	-.396	.293	-.329
.133	-.267	.396	-.369	.404	-.393	.407	-.363	.397	-.343	.494	-.253
.272	-.228	.514	-.304	.497	-.334	.502	-.325	.495	-.316	.590	-.242
.416	-.186	.618	-.259	.599	-.296	.601	-.303	.594	-.295	.693	-.242
.565	-.151	.733	-.196	.700	-.273	.698	-.295	.693	-.293	.777	-.240
.713	-.138	.835	-.155	.864	-.213	.863	-.253	.784	-.277	.861	-.220
.854	-.135	.919	-.089	.926	-.138	.923	-.174	.856	-.258	.918	-.187
.980	-.130	.987	-.029	.975	-.054	.977	-.056	.926	-.173	.972	-.073
1.074	-.095							.977	-.052		
1.122	-.050										
				LOWER SURFACE							
-.660	.023	-.022	.378	.024	.490	.025	.482	.019	.487	.020	.467
-.616	.080	.038	.266	.075	.283	.074	.244	.066	.263	.076	.199
-.572	.087	.101	.174	.297	.060	.130	.174	.136	.149	.136	.080
-.462	.069	.185	.093	.400	.046	.298	.064	.214	.101	.221	.054
-.329	.034	.398	.026	.604	.002	.397	.043	.292	.082	.295	.028
-.172	.004	.737	.137	.785	.188	.501	.023	.403	.050	.396	.011
-.030	-.032			.967	.161	.603	.069	.489	.021	.497	-.005
.128	-.036			1.000	-.008	.703	.105	.594	.025	.597	.006
.418	.014					.784	.170	.700	.103	.702	.092
.564	.048					.868	.250	.786	.187	.786	.156
.710	.116					.923	.283	.858	.255	.864	.208
.976	.232					.972	.175	.919	.269	.912	.214
1.110	.134							.967	.143	.985	.038
CN=	.5650		.5600		.5772		.5782		.5521		.4508
CM=	.0678		-.0294		-.0514		-.0698		-.0691		-.0534

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(a) M = 0.25. Continued.

$\alpha = 7.01^{\circ}$; $C_L = 0.650$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
				UPPER SURFACE							
-.660	-.349	-.021	-1.712	.023	-2.088	.025	-1.953	.022	-1.945	.018	-1.629
-.567	-.477	.035	-1.069	.068	-1.161	.079	-1.114	.075	-1.039	.077	-.932
-.452	-.471	.105	-.807	.134	-.822	.133	-.782	.129	-.749	.129	-.641
-.311	-.460	.178	-.650	.209	-.619	.214	-.592	.201	-.584	.209	-.475
-.023	-.375	.286	-.520	.294	-.506	.295	-.474	.294	-.458	.293	-.380
.133	-.293	.396	-.414	.404	-.418	.407	-.402	.397	-.379	.494	-.277
.272	-.240	.514	-.327	.497	-.361	.502	-.359	.495	-.340	.590	-.273
.416	-.200	.618	-.272	.599	-.311	.601	-.339	.594	-.317	.693	-.253
.565	-.158	.733	-.199	.700	-.292	.698	-.307	.693	-.304	.777	-.262
.713	-.150	.835	-.150	.864	-.199	.863	-.246	.784	-.285	.861	-.218
.854	-.151	.919	-.087	.926	-.131	.923	-.167	.856	-.245	.918	-.185
.980	-.131	.987	-.040	.975	-.060	.977	-.035	.926	-.154	.972	-.075
1.074	-.106							.977	-.052		
1.122	-.065										
				LOWER SURFACE							
-.660	.004	-.022	.350	.024	.516	.025	.520	.019	.514	.020	.507
-.616	.080	.038	.314	.075	.333	.074	.326	.066	.331	.076	.279
-.572	.106	.101	.213	.297	.096	.130	.224	.136	.210	.136	.151
-.462	.078	.185	.151	.400	.068	.298	.110	.214	.131	.221	.078
-.329	.058	.358	.050	.604	.015	.397	.074	.292	.112	.295	.060
-.172	.029	.737	.147	.785	.190	.501	.044	.403	.070	.396	.032
-.030	-.002			.967	.165	.603	.088	.489	.044	.497	-.000
.128	-.004			1.000	-.023	.703	.116	.594	.025	.597	.015
.418	.039					.784	.170	.700	.110	.702	.083
.564	.066					.868	.250	.786	.184	.786	.147
.710	.138					.923	.288	.858	.261	.864	.207
.976	.239					.972	.168	.919	.265	.912	.209
1.110	.132							.967	.136	.985	.021
CN=	.6480		.6380		.6497		.6667		.6309		.5229
CM=	.0889		-.0257		-.0470		-.0643		-.0621		-.0478

$\alpha = 7.99^{\circ}$; $C_L = 0.728$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
				UPPER SURFACE							
-.660	-.397	-.021	-2.050	.023	-2.329	.025	-1.791	.022	-2.452	.018	-2.137
-.567	-.513	.035	-1.231	.068	-1.300	.079	-1.226	.075	-1.172	.077	-1.054
-.452	-.505	.105	-.888	.134	-.894	.133	-.868	.129	-.838	.129	-.727
-.311	-.485	.178	-.727	.209	-.687	.214	-.657	.201	-.651	.209	-.526
-.023	-.392	.286	-.557	.294	-.521	.295	-.530	.294	-.504	.293	-.433
.133	-.308	.396	-.445	.404	-.438	.407	-.439	.397	-.421	.494	-.300
.272	-.266	.514	-.337	.497	-.378	.502	-.386	.495	-.377	.590	-.281
.416	-.220	.618	-.273	.599	-.321	.601	-.342	.594	-.336	.693	-.282
.565	-.184	.733	-.201	.700	-.288	.698	-.325	.693	-.313	.777	-.266
.713	-.164	.835	-.145	.864	-.180	.863	-.247	.784	-.282	.861	-.226
.854	-.160	.919	-.085	.926	-.117	.923	-.153	.856	-.233	.918	-.180
.980	-.138	.987	-.058	.975	-.058	.977	-.051	.926	-.149	.972	-.079
1.074	-.108							.977	-.052		
1.122	-.062										
				LOWER SURFACE							
-.660	-.012	-.022	.310	.024	.538	.025	.542	.019	.527	.020	.521
-.616	.093	.038	.344	.075	.375	.074	.381	.066	.381	.076	.323
-.572	.115	.101	.261	.297	.136	.130	.270	.136	.259	.136	.207
-.462	.098	.185	.180	.400	.098	.298	.131	.214	.173	.221	.119
-.329	.074	.358	.074	.604	.025	.397	.100	.292	.143	.295	.075
-.172	.050	.737	.153	.785	.195	.501	.068	.403	.094	.396	.052
-.030	.017			.967	.155	.603	.101	.489	.062	.497	.017
.128	.021			1.000	-.031	.703	.123	.594	.051	.597	.023
.418	.050					.784	.178	.700	.117	.702	.081
.564	.079					.868	.258	.786	.188	.786	.159
.710	.147					.923	.297	.858	.261	.864	.199
.976	.245					.972	.165	.919	.263	.912	.205
1.110	.141							.967	.128	.985	.007
CN=	.7093		.7051		.7010		.7106		.7080		.5950
CM=	.1005		-.0209		-.0415		-.0685		-.0569		-.0426

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; δ_h = -2.5°

(a) M = 0.25. Continued.

α = 8.93°; C_L = 0.800

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	-.432	-.021	-2.597	.023	-2.731	.025	-1.910	.022	-3.030	.018	-2.597
	-.567	-.564	.035	-1.372	.068	-1.428	.079	-1.411	.075	-1.311	.077	-1.173
	-.452	-.544	.105	-.965	.134	-.970	.133	-.948	.129	-.948	.129	-.813
	-.311	-.510	.178	-.771	.209	-.745	.214	-.698	.201	-.719	.209	-.589
	-.023	-.403	.286	-.580	.294	-.584	.295	-.566	.294	-.557	.293	-.461
	.133	-.332	.396	-.458	.404	-.464	.407	-.461	.397	-.449	.494	-.322
	.272	-.281	.514	-.361	.497	-.386	.502	-.410	.495	-.390	.590	-.294
	.416	-.231	.618	-.283	.599	-.325	.601	-.355	.594	-.344	.693	-.277
	.565	-.196	.733	-.214	.700	-.271	.698	-.324	.693	-.306	.777	-.257
	.713	-.166	.835	-.146	.864	-.162	.863	-.232	.784	-.275	.861	-.211
	.854	-.181	.919	-.100	.926	-.112	.923	-.152	.856	-.228	.918	-.165
	.980	-.134	.987	-.076	.975	-.062	.977	-.064	.926	-.136	.972	-.074
	1.074	-.112							.977	-.058		
	1.122	-.069										
						LOWER SURFACE						
	-.660	-.036	-.022	.268	.024	.523	.025	.564	.019	.512	.020	.514
	-.616	.095	.038	.367	.075	.421	.074	.417	.066	.422	.076	.366
	-.572	.129	.101	.297	.297	.169	.130	.300	.136	.301	.136	.238
	-.462	.125	.185	.220	.400	.125	.298	.157	.214	.214	.221	.145
	-.329	.103	.398	.110	.604	.043	.397	.134	.292	.165	.295	.100
	-.172	.077	.737	.163	.785	.198	.501	.091	.403	.129	.396	.063
	-.030	.043			.967	.150	.603	.122	.489	.084	.497	.029
	.128	.043			1.000	-.047	.703	.133	.594	.068	.597	.033
	.418	.076					.784	.179	.700	.127	.702	.091
	.564	.110					.868	.266	.786	.192	.786	.155
	.710	.166					.923	.298	.858	.262	.864	.194
	.976	.260					.972	.167	.919	.263	.912	.200
	1.110	.153							.967	.124	.985	-.006
CN=	.7812		.7850		.7608		.7631		.7824		.6508	
CM=	.1124		-.0176		-.0351		-.0675		-.0496		-.0344	

α = 9.59°; C_L = 0.851

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	-.519	-.021	-2.733	.023	-2.809	.025	-2.059	.022	-3.278	.018	-2.815
	-.567	-.621	.035	-1.468	.063	-1.572	.079	-1.505	.075	-1.419	.077	-1.254
	-.452	-.575	.105	-1.044	.134	-1.031	.133	-1.034	.129	-1.000	.129	-.875
	-.311	-.550	.178	-.799	.209	-.785	.214	-.752	.201	-.766	.209	-.624
	-.023	-.425	.286	-.622	.294	-.599	.295	-.600	.294	-.592	.293	-.498
	.133	-.337	.396	-.493	.404	-.490	.407	-.492	.397	-.470	.494	-.338
	.272	-.293	.514	-.370	.497	-.400	.502	-.416	.495	-.406	.590	-.307
	.416	-.240	.618	-.288	.599	-.328	.601	-.360	.594	-.360	.693	-.277
	.565	-.210	.733	-.220	.700	-.273	.698	-.328	.693	-.319	.777	-.254
	.713	-.182	.835	-.161	.864	-.166	.863	-.237	.784	-.264	.861	-.202
	.854	-.182	.919	-.121	.926	-.102	.923	-.140	.856	-.210	.918	-.152
	.980	-.144	.987	-.070	.975	-.082	.977	-.066	.926	-.126	.972	-.083
	1.074	-.109							.977	-.066		
	1.122	-.050										
						LOWER SURFACE						
	-.660	-.055	-.022	.224	.024	.510	.025	.536	.019	.496	.020	.503
	-.616	.096	.038	.394	.075	.455	.074	.451	.066	.454	.076	.395
	-.572	.139	.101	.316	.297	.173	.130	.339	.136	.325	.136	.252
	-.462	.140	.185	.245	.400	.143	.298	.187	.214	.238	.221	.172
	-.329	.116	.398	.131	.604	.055	.397	.141	.292	.184	.295	.113
	-.172	.095	.737	.172	.785	.204	.501	.100	.403	.145	.396	.080
	-.030	.054			.967	.142	.603	.125	.489	.097	.497	.035
	.128	.060			1.000	-.054	.703	.129	.594	.059	.597	.034
	.418	.089					.784	.176	.700	.133	.702	.085
	.564	.110					.868	.267	.786	.201	.786	.139
	.710	.178					.923	.303	.858	.272	.864	.193
	.976	.277					.972	.167	.919	.264	.912	.190
	1.110	.155							.967	.122	.985	-.023
CN=	.8388		.8352		.7962		.8047		.8253		.6849	
CM=	.1286		-.0193		-.0344		-.0652		-.0466		-.0294	

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; δ_h = -2.5°

(a) M = 0.25. Continued.

α = 10.49°; C_L = 0.921

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.559	-.021	-3.259	.023	-2.746	.025	-2.350	.022	-3.579	.018	-1.944
-.567	-.658	.035	-1.594	.068	-1.692	.079	-1.633	.075	-1.535	.077	-1.576
-.452	-.619	.105	-1.138	.134	-1.103	.133	-1.085	.129	-1.084	.129	-.988
-.311	-.570	.178	-.885	.209	-.823	.214	-.824	.201	-.826	.209	-.789
-.023	-.462	.286	-.678	.294	-.646	.295	-.632	.294	-.644	.293	-.921
.133	-.368	.396	-.518	.404	-.510	.407	-.509	.397	-.514	.494	-.314
.272	-.314	.514	-.407	.497	-.416	.502	-.432	.495	-.447	.590	-.275
.416	-.271	.618	-.322	.599	-.332	.601	-.373	.594	-.376	.693	-.248
.565	-.213	.733	-.246	.700	-.282	.698	-.322	.693	-.318	.777	-.252
.713	-.190	.835	-.187	.864	-.145	.863	-.219	.784	-.274	.861	-.184
.854	-.186	.919	-.142	.926	-.112	.923	-.134	.856	-.198	.918	-.149
.980	-.155	.987	-.088	.975	-.093	.977	-.067	.926	-.122	.972	-.107
1.074	-.116							.977	-.066		
1.122	-.064										
				LOWER SURFACE							
-.660	-.082	-.022	.127	.024	.477	.025	.524	.019	.439	.020	.511
-.616	.087	.038	.417	.075	.474	.074	.478	.066	.471	.076	.408
-.572	.143	.101	.373	.297	.217	.130	.375	.136	.345	.136	.285
-.462	.160	.185	.281	.400	.138	.298	.221	.214	.262	.221	.180
-.329	.136	.398	.156	.604	.064	.397	.170	.292	.225	.295	.137
-.172	.123	.737	.179	.785	.204	.501	.119	.403	.161	.396	.089
-.030	.089			.967	.127	.603	.138	.489	.111	.497	.043
.128	.091			1.000	-.085	.703	.137	.594	.074	.597	.039
.418	.115					.784	.182	.700	.137	.702	.089
.564	.133					.868	.267	.786	.198	.786	.142
.710	.189					.923	.308	.858	.265	.864	.196
.976	.280					.972	.159	.919	.261	.912	.191
1.110	.169							.967	.114	.985	-.023
CN=	.9130		.9250		.8242		.8573		.8805		.7419
CM=	.1414		-.0193		-.0336		-.0609		-.0442		-.0345

α = 11.51°; C_L = 0.996

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.650	-.617	-.021	-3.527	.023	-2.853	.025	-2.596	.022	-2.183	.018	-.759
-.567	-.743	.035	-1.779	.068	-1.787	.079	-1.724	.075	-1.899	.077	-.711
-.452	-.664	.105	-1.223	.134	-1.176	.133	-1.183	.129	-2.278	.129	-.724
-.311	-.623	.178	-.927	.209	-.892	.214	-.890	.201	-1.759	.209	-.757
-.023	-.476	.286	-.716	.294	-.697	.295	-.695	.294	-1.374	.293	-.710
.133	-.387	.396	-.567	.404	-.536	.407	-.549	.397	-.613	.494	-.567
.272	-.339	.514	-.447	.497	-.433	.502	-.448	.495	-.433	.590	-.520
.416	-.295	.618	-.349	.599	-.337	.601	-.377	.594	-.281	.693	-.448
.565	-.248	.733	-.267	.700	-.267	.698	-.328	.693	-.261	.777	-.428
.713	-.199	.835	-.203	.864	-.163	.863	-.187	.784	-.225	.861	-.380
.854	-.200	.919	-.147	.926	-.157	.923	-.141	.856	-.195	.918	-.330
.980	-.161	.987	-.100	.975	-.124	.977	-.107	.926	-.167	.972	-.343
1.074	-.107							.977	-.067		
1.122	-.060										
				LOWER SURFACE							
-.660	-.097	-.022	.021	.024	.419	.025	.490	.019	.381	.020	.519
-.616	.085	.038	.429	.075	.511	.074	.512	.066	.496	.076	.410
-.572	.154	.101	.388	.297	.250	.130	.424	.136	.390	.136	.299
-.462	.162	.185	.315	.400	.192	.298	.239	.214	.301	.221	.190
-.329	.151	.398	.174	.604	.079	.397	.188	.292	.236	.295	.150
-.172	.139	.737	.196	.785	.213	.501	.133	.403	.179	.396	.091
-.030	.116			.967	.114	.603	.150	.489	.132	.497	.048
.128	.111			1.000	-.106	.703	.145	.594	.088	.597	.041
.418	.131					.784	.180	.700	.139	.702	.087
.564	.163					.868	.273	.786	.202	.786	.124
.710	.218					.923	.306	.858	.257	.864	.181
.976	.299					.972	.150	.919	.262	.912	.172
1.110	.181							.967	.100	.985	-.078
CN=	.9941		.9966		.8773		.9117		1.0592		.7172
CM=	.1550		-.0231		-.0372		-.0583		-.0422		-.1106

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25. Continued.

$\alpha = 12.43^o$; $C_L = 1.062$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.650	-.714	-.021	-3.644	.023	-2.964	.025	-2.483	.022	-1.391	.018	-.568
-.567	-.803	.035	-1.913	.068	-1.919	.079	-1.582	.075	-1.484	.077	-.617
-.452	-.730	.105	-1.325	.134	-1.275	.133	-1.197	.129	-1.219	.129	-.620
-.311	-.661	.178	-1.032	.209	-.928	.214	-.907	.201	-1.360	.209	-.557
-.023	-.499	.286	-.772	.294	-.799	.295	-.701	.294	-1.350	.293	-.550
.133	-.430	.396	-.625	.404	-.621	.407	-.529	.397	-1.247	.494	-.445
.272	-.362	.514	-.485	.497	-.490	.502	-.394	.495	-1.048	.590	-.425
.416	-.322	.618	-.397	.599	-.386	.601	-.328	.594	-.467	.693	-.407
.565	-.268	.733	-.304	.700	-.302	.698	-.221	.693	-.630	.777	-.403
.713	-.226	.835	-.228	.864	-.210	.863	-.272	.784	-.250	.861	-.395
.854	-.216	.919	-.149	.926	-.176	.923	-.214	.856	-.391	.918	-.337
.980	-.168	.987	-.106	.975	-.149	.977	-.234	.926	-.255	.972	-.336
1.074	-.100							.977	-.156		
1.122	-.056										
					LOWER SURFACE						
-.660	-.156	-.022	-.136	.024	.359	.025	.446	.019	.432	.020	.518
-.616	.077	.038	.433	.075	.523	.074	.542	.066	.520	.076	.400
-.572	.154	.101	.415	.297	.262	.130	.448	.136	.417	.136	.277
-.462	.191	.185	.338	.400	.198	.298	.262	.214	.321	.221	.195
-.329	.191	.398	.200	.604	.085	.397	.202	.292	.260	.295	.138
-.172	.149	.737	.196	.785	.209	.501	.148	.403	.190	.396	.093
-.030	.139			.967	.100	.603	.166	.489	.139	.497	.036
.128	.121			1.000	-.145	.703	.146	.594	.090	.597	.025
.418	.156					.784	.173	.700	.144	.702	.069
.564	.165					.868	.268	.786	.193	.786	.106
.710	.237					.923	.317	.858	.257	.864	.143
.976	.305					.972	.132	.919	.258	.912	.144
1.110	.177							.967	.097	.985	-.156
CN=	1.0764		1.0637		.9417		.9164		1.1107		.6078
CM=	.1746		-.0287		-.0451		-.0707		-.1259		-.0967

$\alpha = 13.34^o$; $C_L = 1.129$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.792	-.021	-3.895	.023	-3.475	.025	-2.420	.022	-1.246	.018	-.616
-.567	-.842	.035	-2.028	.068	-1.886	.079	-1.522	.075	-1.258	.077	-.597
-.452	-.768	.105	-1.416	.134	-1.961	.133	-1.047	.129	-1.187	.129	-.571
-.311	-.677	.178	-1.084	.209	-1.046	.214	-.841	.201	-1.160	.209	-.574
-.023	-.515	.286	-.864	.294	-.902	.295	-.710	.294	-1.227	.293	-.509
.133	-.450	.396	-.691	.404	-.697	.407	-.558	.397	-1.313	.494	-.450
.272	-.383	.514	-.531	.497	-.593	.502	-.468	.495	-1.087	.590	-.465
.416	-.335	.618	-.426	.599	-.478	.601	-.395	.594	-.888	.693	-.421
.565	-.279	.733	-.313	.700	-.499	.698	-.384	.693	-.737	.777	-.404
.713	-.251	.835	-.215	.864	-.254	.863	-.321	.784	-.628	.861	-.386
.854	-.201	.919	-.148	.926	-.197	.923	-.274	.856	-.507	.918	-.380
.980	-.161	.987	-.109	.975	-.189	.977	-.252	.926	-.404	.972	-.353
1.074	-.112							.977	-.275		
1.122	-.057										
					LOWER SURFACE						
-.660	-.175	-.022	-.266	.024	.304	.025	.431	.019	.463	.020	.507
-.616	.060	.038	.414	.075	.545	.074	.516	.066	.521	.076	.381
-.572	.162	.101	.420	.297	.279	.130	.414	.136	.401	.136	.275
-.462	.177	.185	.351	.400	.199	.298	.240	.214	.310	.221	.168
-.329	.207	.398	.224	.604	.082	.397	.207	.292	.241	.295	.128
-.172	.164	.737	.197	.785	.189	.501	.129	.403	.179	.396	.064
-.030	.159			.967	.074	.603	.146	.489	.121	.497	.016
.128	.123			1.000	-.215	.703	.117	.594	.083	.597	-.011
.418	.166					.784	.149	.700	.130	.702	.019
.564	.177					.868	.245	.786	.191	.786	.072
.710	.241					.923	.282	.858	.258	.864	.117
.976	.300					.972	.085	.919	.259	.912	.118
1.110	.174							.967	.103	.985	-.227
CN=	1.1168		1.1223		1.0881		.9072		1.1702		.5822
CM=	.1878		-.0293		-.0545		-.0814		-.1812		-.0903

[redacted]

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TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(a) M = 0.25. Concluded.

$\alpha = 14.13^{\circ}$; $C_L = 1.171$

	STA .133		STA .307		STA .480		STA .653		STA .804		STA .933	
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.853	-.021	-4.139	.023	-1.354	.025	-2.282	.022	-1.299	.018	-.573
	-.567	-.908	.035	-2.161	.068	-1.223	.079	-1.140	.075	-1.310	.077	-.553
	-.452	-.816	.105	-1.471	.134	-1.306	.133	-.809	.129	-1.355	.129	-.542
	-.311	-.713	.178	-1.308	.209	-1.236	.214	-.752	.201	-1.434	.209	-.546
	-.023	-.546	.286	-.954	.294	-1.137	.295	-.620	.294	-1.490	.293	-.543
	.133	-.506	.396	-.774	.404	-1.020	.407	-.457	.397	-1.341	.494	-.472
	.272	-.432	.514	-.569	.497	-.961	.502	-.515	.495	-1.137	.590	-.460
	.416	-.375	.618	-.456	.599	-.900	.601	-.439	.594	-.802	.693	-.411
	.565	-.312	.733	-.336	.700	-.796	.698	-.350	.693	-.487	.777	-.434
	.713	-.275	.835	-.258	.864	-.630	.863	-.422	.784	-.488	.861	-.417
	.854	-.243	.919	-.199	.926	-.586	.923	-.365	.856	-.372	.918	-.412
	.980	-.181	.987	-.152	.975	-.472	.977	-.303	.926	-.265	.972	-.398
	1.074	-.137							.977	-.176		
	1.122	-.065										
LOWER SURFACE												
	-.660	-.211	-.022	-.383	.024	.512	.025	.460	.019	.466	.020	.526
	-.616	.073	.038	.452	.075	.530	.074	.491	.066	.509	.076	.431
	-.572	.169	.101	.485	.297	.268	.130	.391	.136	.394	.136	.299
	-.462	.229	.185	.395	.400	.210	.298	.253	.214	.311	.221	.184
	-.329	.217	.398	.245	.604	.053	.397	.190	.292	.259	.295	.137
	-.172	.207	.737	.186	.785	.142	.501	.120	.403	.172	.396	.087
	-.030	.167			.967	-.065	.603	.123	.489	.137	.497	.032
	.128	.162			1.000	-.448	.703	.086	.594	.081	.597	.003
	.418	.197					.784	.130	.700	.127	.702	.052
	.564	.193					.868	.215	.786	.207	.786	.076
	.710	.264					.923	.259	.858	.284	.864	.124
	.976	.325					.972	.028	.919	.254	.912	.121
	1.110	.183							.967	.129	.985	-.235
CN=	1.2312		1.2185		1.1530		.8442		1.1808		.6056	
CM=	.1974		-.0327		-.1662		-.0923		-.1488		-.0995	

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TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50.

$\alpha = -5.24^{\circ}$; $C_L = -0.459$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.051	-.021	.343	.023	.434	.025	.436	.022	.506	.018	.504
-.567	.006	.035	.200	.068	.304	.079	.284	.075	.314	.077	.258
-.452	-.049	.105	.118	.209	.111	.133	.209	.129	.245	.129	.204
-.311	-.099	.178	.065	.294	.064	.214	.154	.201	.173	.209	.133
-.023	-.069	.286	.016	.404	.028	.295	.099	.294	.120	.293	.081
.133	-.022	.396	.003	.497	-.002	.407	.059	.397	.078	.494	-.007
.272	.003	.514	-.003	.599	-.030	.502	.010	.495	.023	.590	-.048
.416	.024	.618	-.026	.700	-.071	.601	-.030	.594	-.019	.693	-.100
.565	.026	.733	-.038	.864	-.124	.698	-.082	.693	-.075	.777	-.139
.713	.030	.835	-.066	.926	-.113	.863	-.150	.784	-.123	.861	-.165
.854	.004	.919	-.063			.923	-.139	.856	-.158	.918	-.205
.980	-.017	.987	-.039			.977	-.060	.926	-.156	.972	-.224
1.074	-.033							.977	-.064		
				LOWER SURFACE							
-.660	-.075	-.022	-1.223	.024	-2.022	.074	-1.119	.019	-1.084	.020	-.585
-.616	-.166	.038	-.885	.075	-1.027	.130	-.814	.066	-1.285	.076	-.585
-.462	-.237	.101	-.744	.297	-.458	.298	-.463	.136	-1.117	.136	-.457
-.329	-.247	.185	-.572	.400	-.370	.397	-.361	.214	-1.122	.221	-.500
-.172	-.265	.398	-.358	.604	-.218	.501	-.274	.292	-.938	.295	-.426
-.030	-.323	.737	-.004	.785	.048	.603	-.119	.403	-.342	.396	-.356
.128	-.354			.967	.150	.784	.052	.489	-.467	.497	-.294
.413	-.250			1.000	.059	.868	.125	.594	-.252	.597	-.270
.564	-.189					.923	.130	.700	-.023	.702	-.224
.710	-.062					.972	.094	.786	.056	.786	-.212
.976	.131							.858	.147	.864	-.165
1.072	.131							.919	.175	.912	-.174
1.110	.107							.967	.134	.985	-.169
CN=	-.2807		-.4085		-.4154		-.3621		-.5058		-.3455
CM=	-.0890		-.0475		-.0673		-.0583		-.0610		-.0303

$\alpha = -4.18^{\circ}$; $C_L = -0.359$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.048	-.021	.296	.023	.359	.025	.380	.022	.473	.018	.506
-.567	-.021	.035	.139	.068	.224	.079	.227	.075	.265	.077	.244
-.452	-.083	.105	.072	.209	.054	.133	.160	.129	.193	.129	.184
-.311	-.129	.178	.013	.294	.024	.214	.090	.201	.131	.209	.115
-.023	-.098	.286	-.024	.404	-.002	.295	.070	.294	.089	.293	.058
.133	-.043	.396	-.038	.497	-.023	.407	.017	.397	.046	.494	-.017
.272	-.018	.514	-.033	.599	-.059	.502	-.016	.495	-.005	.590	-.050
.416	.005	.618	-.049	.700	-.090	.601	-.050	.594	-.038	.693	-.097
.565	.015	.733	-.064	.864	-.135	.698	-.094	.693	-.092	.777	-.127
.713	.022	.835	-.080	.926	-.119	.863	-.160	.784	-.129	.861	-.134
.854	-.013	.919	-.079			.923	-.142	.856	-.162	.918	-.139
.980	-.024	.987	-.037			.977	-.043	.926	-.153	.972	-.063
1.074	-.036							.977	-.048		
				LOWER SURFACE							
-.660	-.044	-.022	-.913	.024	-1.426	.074	-.939	.019	-2.257	.020	-1.042
-.616	-.123	.038	-.711	.075	-.860	.130	-.689	.066	-1.075	.076	-1.136
-.462	-.203	.101	-.607	.297	-.406	.298	-.389	.136	-.709	.136	-.769
-.329	-.211	.185	-.489	.400	-.320	.397	-.324	.214	-.513	.221	-.731
-.172	-.238	.398	-.320	.604	-.196	.501	-.234	.292	-.400	.295	-.349
-.030	-.284	.737	.010	.785	.060	.603	-.097	.403	-.310	.396	-.254
.128	-.322			.967	.169	.784	.064	.489	-.258	.497	-.174
.418	-.226			1.000	.048	.868	.144	.594	-.169	.597	-.098
.564	-.168					.923	.158	.700	-.039	.702	.002
.710	-.043					.972	.125	.786	.033	.786	.041
.976	.136							.858	.102	.864	.090
1.072	.137							.919	.121	.912	.103
1.110	.108							.967	.116	.985	.054
CN=	-.1991		-.3060		-.3054		-.2749		-.3704		-.3148
CM=	-.0709		-.0463		-.0648		-.0617		-.0703		-.0644

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{o}$

(b) M = 0.50. Continued.

$\alpha = -3.18^{o}$; $C_L = -0.267$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.029	-.021	.260	.023	.261	.025	.299	.022	.434	.018	.463
-.567	-.038	.035	.080	.068	.135	.079	.147	.075	.183	.077	.197
-.452	-.115	.105	-.012	.209	.004	.133	.093	.129	.133	.129	.137
-.311	-.153	.178	-.032	.294	-.020	.214	.042	.201	.074	.209	.076
-.023	-.127	.286	-.074	.404	-.045	.295	.015	.294	.045	.293	.033
.133	-.070	.396	-.072	.497	-.067	.407	-.020	.397	.003	.494	-.040
.272	-.043	.514	-.065	.599	-.087	.502	-.050	.495	-.036	.590	-.074
.416	-.025	.618	-.076	.700	-.122	.601	-.082	.594	-.067	.693	-.110
.565	-.007	.733	-.077	.864	-.154	.698	-.124	.693	-.115	.777	-.139
.713	-.002	.835	-.095	.926	-.131	.863	-.177	.784	-.149	.861	-.138
.854	-.016	.919	-.089			.923	-.148	.856	-.171	.918	-.138
.980	-.042	.987	-.042			.977	-.052	.926	-.155	.972	-.048
1.074	-.042							.977	-.047		
				LOWER SURFACE							
-.660	-.026	-.022	-.688	.024	-1.138	.074	-.741	.019	-1.695	.020	-1.909
-.616	-.106	.038	-.578	.075	-.721	.130	-.572	.066	-.911	.076	-.909
-.462	-.169	.101	-.533	.297	-.364	.298	-.346	.136	-.605	.136	-.645
-.329	-.184	.165	-.407	.400	-.295	.397	-.294	.214	-.445	.221	-.431
-.172	-.231	.398	-.293	.604	-.179	.501	-.222	.292	-.346	.295	-.322
-.030	-.263	.737	.031	.785	.074	.603	-.093	.403	-.287	.396	-.235
.128	-.310			.967	.175	.784	.077	.489	-.234	.497	-.167
.418	-.212			1.000	.042	.868	.153	.594	-.160	.597	-.068
.564	-.142					.923	.171	.700	-.010	.702	.032
.710	-.030					.972	.131	.786	.075	.786	.072
.976	.146							.858	.116	.864	.090
1.072	.137							.919	.160	.912	.116
1.110	.103							.967	.132	.985	.071
CN=	-.1296		-.2132		-.2138		-.1885		-.2694		-.2734
CM=	-.0606		-.0480		-.0666		-.0636		-.0721		-.0763

$\alpha = -2.14^{o}$; $C_L = -0.170$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.022	-.021	.199	.023	.171	.025	.183	.022	.331	.018	.407
-.567	-.001	.035	-.024	.068	.044	.079	.070	.075	.091	.077	.101
-.452	-.143	.105	-.066	.209	-.057	.133	.022	.129	.068	.129	.080
-.311	-.179	.178	-.083	.294	-.066	.214	-.007	.201	.023	.209	.030
-.023	-.144	.286	-.108	.404	-.075	.295	-.038	.294	-.001	.293	-.012
.133	-.096	.396	-.109	.497	-.101	.407	-.060	.397	-.026	.494	-.061
.272	-.069	.514	-.092	.599	-.114	.502	-.082	.495	-.064	.590	-.090
.416	-.042	.618	-.102	.700	-.137	.601	-.114	.594	-.095	.693	-.124
.565	-.027	.733	-.098	.864	-.163	.698	-.142	.693	-.133	.777	-.153
.713	-.018	.835	-.106	.926	-.136	.863	-.192	.784	-.168	.861	-.141
.854	-.034	.919	-.096			.923	-.160	.856	-.191	.918	-.141
.980	-.041	.987	-.045			.977	-.052	.926	-.164	.972	-.046
1.074	-.047							.977	-.047		
				LOWER SURFACE							
-.660	-.002	-.022	-.421	.024	-.803	.074	-.639	.019	-1.217	.020	-1.157
-.616	-.076	.038	-.481	.075	-.611	.130	-.466	.066	-.712	.076	-.773
-.462	-.148	.101	-.445	.297	-.307	.298	-.304	.136	-.513	.136	-.535
-.329	-.163	.185	-.352	.400	-.257	.397	-.254	.214	-.381	.221	-.358
-.172	-.204	.398	-.243	.604	-.155	.501	-.188	.292	-.302	.295	-.270
-.030	-.231	.737	.046	.785	.100	.603	-.081	.403	-.247	.396	-.209
.128	-.266			.967	.172	.784	.103	.489	-.214	.497	-.148
.418	-.181			1.000	.038	.868	.162	.594	-.147	.597	-.070
.564	-.125					.923	.183	.700	.002	.702	.056
.710	-.018					.972	.138	.786	.103	.786	.120
.976	.154							.858	.156	.864	.153
1.072	.142							.919	.181	.912	.155
1.110	.107							.967	.139	.985	.085
CN=	-.1206		-.1206		-.1217		-.1069		-.1701		-.1544
CM=	-.0487		-.0486		-.0673		-.0682		-.0726		-.0750

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = -1.19^{\circ}$; $C_L = -0.082$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE											
-.660	.009	-.021	.091	.023	.001	.025	.098	.022	.231	.018	.333
-.567	-.116	.035	-.102	.068	-.048	.079	-.030	.075	.020	.077	.017
-.452	-.167	.105	-.134	.209	-.105	.133	-.042	.129	.001	.129	.019
-.311	-.212	.178	-.161	.294	-.118	.214	-.064	.201	-.029	.209	-.010
-.023	-.187	.286	-.166	.404	-.120	.295	-.076	.294	-.045	.293	-.044
.133	-.116	.396	-.143	.497	-.133	.407	-.092	.397	-.062	.494	-.084
.272	-.087	.514	-.121	.599	-.142	.502	-.111	.495	-.099	.590	-.107
.416	-.064	.618	-.123	.700	-.166	.601	-.131	.594	-.123	.693	-.138
.565	-.039	.733	-.110	.864	-.180	.698	-.167	.693	-.156	.777	-.166
.713	-.033	.835	-.123	.926	-.146	.863	-.209	.784	-.184	.861	-.160
.854	-.051	.919	-.106			.923	-.162	.856	-.201	.918	-.147
.980	-.060	.967	-.047			.977	-.049	.926	-.171	.972	-.047
1.074	-.058							.977	-.047		
LOWER SURFACE											
-.660	.015	-.022	-.253	.024	-.551	.074	-.552	.019	-1.018	.020	-.833
-.616	-.046	.038	-.362	.075	-.502	.130	-.372	.066	-.678	.076	-.628
-.462	-.124	.101	-.361	.297	-.260	.298	-.255	.136	-.384	.136	-.495
-.329	-.140	.185	-.319	.400	-.221	.397	-.214	.214	-.307	.221	-.291
-.172	-.172	.398	-.225	.604	-.140	.501	-.168	.292	-.254	.295	-.236
-.030	-.221	.737	.060	.785	.118	.603	-.070	.403	-.215	.396	-.189
.128	-.248			.967	.172	.784	.118	.489	-.198	.497	-.137
.418	-.150			1.000	.030	.868	.161	.594	-.130	.597	-.064
.564	-.103					.923	.189	.700	.018	.702	.064
.710	-.001					.972	.141	.786	.120	.786	.141
.976	.157							.858	.177	.864	.174
1.072	.142							.919	.200	.912	.187
1.110	.107							.967	.143	.985	.082
CN=	.0251		-.0405		-.0307		-.0357		-.0920		-.0903
CM=	-.0341		-.0481		-.0667		-.0707		-.0763		-.0753

$\alpha = 0^{\circ}$; $C_L = 0.029$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE											
-.660	-.021	-.021	-.058	.023	-.158	.025	-.070	.022	.046	.018	.187
-.567	-.146	.035	-.227	.068	-.182	.079	-.154	.075	-.106	.077	-.064
-.452	-.199	.105	-.236	.209	-.181	.133	-.133	.129	-.094	.129	-.056
-.311	-.243	.178	-.221	.294	-.171	.214	-.136	.201	-.096	.209	-.062
-.023	-.203	.286	-.208	.404	-.162	.295	-.139	.294	-.103	.293	-.087
.133	-.139	.396	-.180	.497	-.162	.407	-.140	.397	-.104	.494	-.111
.272	-.113	.514	-.152	.599	-.164	.502	-.147	.495	-.135	.590	-.130
.416	-.088	.618	-.155	.700	-.186	.601	-.166	.594	-.147	.693	-.158
.565	-.058	.733	-.135	.864	-.197	.698	-.194	.693	-.184	.777	-.178
.713	-.044	.835	-.133	.926	-.150	.863	-.215	.784	-.212	.861	-.172
.854	-.062	.919	-.108			.923	-.167	.856	-.220	.918	-.163
.980	-.064	.987	-.048			.977	-.052	.926	-.179	.972	-.059
1.074	-.063							.977	-.051		
LOWER SURFACE											
-.660	.027	-.022	-.039	.024	-.246	.074	-.365	.019	-.545	.020	-.489
-.616	-.034	.038	-.218	.075	-.311	.130	-.310	.066	-.447	.076	-.459
-.462	-.093	.101	-.250	.297	-.220	.298	-.191	.136	-.337	.136	-.369
-.329	-.110	.185	-.231	.400	-.168	.397	-.170	.214	-.231	.221	-.226
-.172	-.141	.398	-.178	.604	-.116	.501	-.141	.292	-.197	.295	-.181
-.030	-.193	.737	.071	.785	.135	.603	-.055	.403	-.161	.396	-.151
.128	-.215			.967	.171	.784	.130	.489	-.157	.497	-.120
.418	-.130			1.000	.024	.868	.183	.594	-.108	.597	-.058
.564	-.083					.923	.208	.700	.034	.702	.068
.710	.019					.972	.159	.786	.141	.786	.153
.976	.168							.858	.206	.864	.202
1.072	.151							.919	.227	.912	.208
1.110	.112							.967	.146	.985	.071
CN=	.1029		.0657		.0792		.0632		.0251		.0033
CM=	-.0207		-.0455		-.0666		-.0735		-.0772		-.0740

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(b) M = 0.50. Continued.

$\alpha = 0.97^o$; $C_L = 0.117$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	-.048	-.021	-.219	.023	-.313	.025	-.214	.022	-.088	.018	.015
-.567	-.206	.035	-.299	.068	-.293	.079	-.229	.075	-.222	.077	-.147
-.452	-.239	.105	-.308	.209	-.247	.133	-.211	.129	-.172	.129	-.135
-.311	-.265	.178	-.293	.294	-.210	.214	-.187	.201	-.154	.209	-.132
-.023	-.233	.286	-.262	.404	-.200	.295	-.179	.294	-.149	.293	-.134
.133	-.163	.396	-.227	.497	-.196	.407	-.179	.397	-.147	.494	-.134
.272	-.132	.514	-.185	.599	-.199	.502	-.179	.495	-.164	.590	-.149
.416	-.105	.618	-.167	.700	-.205	.601	-.190	.594	-.176	.693	-.174
.555	-.064	.733	-.150	.864	-.205	.698	-.221	.693	-.208	.777	-.202
.713	-.062	.835	-.143	.926	-.156	.863	-.227	.784	-.226	.861	-.185
.854	-.073	.919	-.114			.923	-.179	.856	-.229	.918	-.176
.980	-.079	.987	-.050			.977	-.057	.926	-.186	.972	-.068
1.074	-.067							.977	-.055		
LOWER SURFACE											
-.660	.031	-.022	.079	.024	-.146	.074	-.235	.019	-.260	.020	-.261
-.616	.003	.038	-.132	.075	-.222	.130	-.213	.066	-.313	.076	-.335
-.462	-.077	.101	-.176	.297	-.176	.298	-.149	.136	-.231	.136	-.300
-.329	-.093	.185	-.161	.400	-.135	.397	-.133	.214	-.185	.221	-.227
-.172	-.131	.398	-.146	.604	-.106	.501	-.108	.292	-.157	.295	-.149
-.030	-.160	.737	.083	.785	.142	.603	-.034	.403	-.116	.396	-.112
.128	-.197			.967	.169	.784	.136	.489	-.115	.497	-.102
.418	-.109			1.000	.022	.868	.203	.594	-.090	.597	-.045
.564	-.064					.923	.238	.700	.044	.702	.073
.710	.034					.972	.163	.786	.156	.786	.162
.976	.181							.858	.232	.864	.224
1.072	.150							.919	.251	.912	.228
1.110	.112							.967	.150	.985	.062
CN=	.1747		.1572		.1544		.1473		.1175		.0910
CM=	-.0061		-.0447		-.0670		-.0774		-.0791		-.0745

$\alpha = 1.98^o$; $C_L = 0.206$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	-.074	-.021	-.344	.023	-.527	.025	-.374	.022	-.273	.018	-.155
-.567	-.242	.035	-.439	.068	-.400	.079	-.358	.075	-.352	.077	-.257
-.452	-.291	.105	-.376	.209	-.294	.133	-.317	.129	-.239	.129	-.199
-.311	-.303	.178	-.350	.294	-.275	.214	-.248	.201	-.227	.209	-.166
-.023	-.253	.286	-.290	.404	-.226	.295	-.227	.294	-.192	.293	-.171
.133	-.197	.396	-.254	.497	-.229	.407	-.215	.397	-.192	.494	-.150
.272	-.150	.514	-.207	.599	-.221	.502	-.209	.495	-.192	.590	-.171
.416	-.123	.618	-.196	.700	-.226	.601	-.216	.594	-.200	.693	-.193
.565	-.084	.733	-.172	.864	-.213	.698	-.240	.693	-.227	.777	-.209
.713	-.075	.835	-.157	.926	-.159	.863	-.237	.784	-.241	.861	-.198
.854	-.084	.919	-.119			.923	-.177	.856	-.246	.918	-.177
.980	-.082	.987	-.044			.977	-.055	.926	-.190	.972	-.066
1.074	-.071							.977	-.059		
LOWER SURFACE											
-.660	.044	-.022	.182	.024	.069	.074	-.133	.019	-.041	.020	-.025
-.616	.010	.038	-.029	.075	-.130	.130	-.128	.066	-.164	.076	-.200
-.462	-.047	.101	-.094	.297	-.126	.298	-.120	.136	-.158	.136	-.215
-.329	-.072	.185	-.126	.400	-.114	.397	-.098	.214	-.142	.221	-.149
-.172	-.099	.398	-.109	.604	-.080	.501	-.082	.292	-.112	.295	-.143
-.030	-.143	.737	.092	.785	.153	.603	-.022	.403	-.091	.396	-.086
.128	-.168			.967	.176	.784	.142	.489	-.099	.497	-.076
.418	-.087			1.000	.018	.868	.218	.594	-.071	.597	-.035
.564	-.041					.923	.256	.700	.057	.702	.071
.710	.054					.972	.170	.786	.161	.786	.166
.976	.188							.858	.237	.864	.223
1.072	.163							.919	.255	.912	.231
1.110	.120							.967	.144	.985	.056
CN=	.2540		.2363		.2398		.2268		.2013		.1536
CM=	.0093		-.0440		-.0668		-.0784		-.0779		-.0707

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = 2.48^{\circ}$; $C_L = 0.252$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.098	-.021	-.502	.023	-.698	.025	-.513	.022	-.412	.018	-.260
-.557	-.251	.035	-.492	.068	-.466	.079	-.423	.075	-.400	.077	-.310
-.452	-.285	.105	-.418	.209	-.317	.133	-.337	.129	-.303	.129	-.257
-.311	-.313	.178	-.377	.294	-.296	.214	-.281	.201	-.258	.209	-.192
-.023	-.265	.286	-.316	.404	-.251	.295	-.249	.294	-.226	.293	-.182
.133	-.194	.396	-.273	.497	-.237	.407	-.239	.397	-.209	.494	-.161
.272	-.163	.514	-.218	.599	-.228	.502	-.224	.495	-.205	.590	-.177
.416	-.125	.618	-.205	.700	-.239	.601	-.235	.594	-.218	.693	-.194
.565	-.096	.733	-.172	.864	-.212	.698	-.242	.693	-.231	.777	-.219
.713	-.080	.835	-.153	.926	-.155	.863	-.237	.784	-.251	.861	-.193
.854	-.088	.919	-.116			.923	-.184	.856	-.244	.918	-.181
.980	-.086	.987	-.039			.977	-.060	.926	-.183	.972	-.069
1.074	-.073							.977	-.055		
				LOWER SURFACE							
-.660	.045	-.022	.233	.024	.132	.074	-.087	.019	.012	.020	.056
-.616	.021	.038	.014	.075	-.035	.130	-.102	.066	-.112	.076	-.163
-.462	-.035	.101	-.070	.297	-.109	.298	-.085	.136	-.132	.136	-.197
-.329	-.063	.185	-.089	.400	-.081	.397	-.089	.214	-.103	.221	-.129
-.172	-.094	.398	-.087	.604	-.074	.501	-.076	.292	-.090	.295	-.109
-.030	-.121	.737	.100	.785	.158	.603	-.012	.403	-.081	.396	-.074
.128	-.157			.967	.172	.784	.140	.489	-.079	.497	-.070
.418	-.071			1.000	.012	.868	.218	.594	-.063	.597	-.027
.564	-.031					.923	.262	.700	.062	.702	.080
.710	.056					.972	.166	.786	.162	.786	.159
.976	.196							.858	.234	.864	.218
1.072	.164							.919	.257	.912	.227
1.110	.118							.967	.143	.985	.055
CN=	.2830		.2626		.2889		.2656		.2422		.1864
CM=	.0128		-.0418		-.0638		-.0778		-.0764		-.0686

$\alpha = 2.90^{\circ}$; $C_L = 0.291$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.114	-.021	-.590	.023	-.760	.025	-.575	.022	-.555	.018	-.371
-.567	-.271	.035	-.557	.068	-.554	.079	-.460	.075	-.476	.077	-.356
-.452	-.305	.105	-.454	.209	-.360	.133	-.380	.129	-.348	.129	-.289
-.311	-.330	.178	-.415	.294	-.317	.214	-.324	.201	-.290	.209	-.223
-.023	-.282	.286	-.333	.404	-.271	.295	-.278	.294	-.243	.293	-.200
.133	-.204	.396	-.294	.497	-.255	.407	-.252	.397	-.222	.494	-.175
.272	-.177	.514	-.234	.599	-.242	.502	-.240	.495	-.219	.590	-.187
.416	-.138	.618	-.212	.700	-.239	.601	-.242	.594	-.230	.693	-.206
.565	-.099	.733	-.179	.864	-.219	.698	-.255	.693	-.247	.777	-.222
.713	-.086	.835	-.155	.926	-.157	.863	-.245	.784	-.248	.861	-.202
.854	-.094	.919	-.115			.923	-.182	.856	-.247	.918	-.186
.980	-.096	.987	-.044			.977	-.062	.926	-.187	.972	-.068
1.074	-.077							.977	-.054		
				LOWER SURFACE							
-.660	.045	-.022	.269	.024	.182	.074	-.028	.019	.112	.020	.146
-.616	.036	.038	.040	.075	-.004	.130	-.045	.066	-.066	.076	-.117
-.462	-.016	.101	-.032	.297	-.089	.298	-.067	.136	-.095	.136	-.145
-.329	-.050	.185	-.068	.400	-.067	.397	-.072	.214	-.071	.221	-.122
-.172	-.080	.398	-.080	.604	-.064	.501	-.061	.292	-.068	.295	-.099
-.030	-.107	.737	.099	.785	.161	.603	-.005	.403	-.063	.396	-.076
.128	-.140			.967	.167	.784	.143	.489	-.068	.497	-.067
.418	-.065			1.000	.010	.868	.227	.594	-.058	.597	-.026
.564	-.020					.923	.269	.700	.063	.702	.074
.710	.068					.972	.168	.786	.169	.786	.157
.976	.201							.858	.243	.864	.221
1.072	.168							.919	.258	.912	.222
1.110	.122							.967	.144	.985	.049
CN=	.3262		.3168		.3265		.3070		.2850		.2182
CM=	.0214		-.0397		-.0632		-.0788		-.0752		-.0663

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = 3.38^{\circ}$; $C_L = 0.337$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	-.135	-.021	-.677	.023	-.892	.025	-.749	.022	-.648	.018	-.507
	-.567	-.289	.035	-.630	.068	-.601	.079	-.534	.075	-.516	.077	-.419
	-.452	-.328	.105	-.503	.209	-.405	.133	-.438	.129	-.392	.129	-.315
	-.311	-.347	.178	-.426	.294	-.344	.214	-.345	.201	-.328	.209	-.256
	-.023	-.289	.286	-.355	.404	-.291	.295	-.303	.294	-.277	.293	-.230
	.133	-.218	.396	-.305	.497	-.268	.407	-.265	.397	-.238	.494	-.192
	.272	-.177	.514	-.243	.599	-.252	.502	-.266	.495	-.232	.590	-.192
	.416	-.140	.619	-.223	.700	-.251	.601	-.255	.594	-.239	.693	-.213
	.555	-.115	.733	-.186	.864	-.217	.698	-.268	.693	-.254	.777	-.230
	.713	-.093	.835	-.157	.926	-.152	.863	-.247	.784	-.262	.861	-.207
	.854	-.100	.919	-.109			.923	-.187	.856	-.255	.918	-.181
	.980	-.094	.967	-.037			.977	-.060	.926	-.188	.972	-.065
	1.074	-.075							.977	-.053		
						LOWER SURFACE						
	-.660	.041	-.022	.304	.024	.265	.074	.010	.019	.169	.020	.194
	-.616	.035	.038	.090	.075	.035	.130	-.022	.066	.017	.076	-.061
	-.462	-.007	.101	-.002	.297	-.062	.258	-.052	.136	-.053	.136	-.082
	-.329	-.036	.185	-.046	.400	-.050	.397	-.045	.214	-.060	.221	-.081
	-.172	-.068	.398	-.072	.604	-.063	.501	-.052	.292	-.053	.295	-.086
	-.030	-.103	.737	.106	.785	.168	.603	.004	.403	-.044	.396	-.067
	.128	-.135			.967	.165	.784	.149	.483	-.055	.497	-.067
	.418	-.049			1.000	.004	.868	.224	.594	-.043	.597	-.022
	.564	-.010					.923	.267	.700	.069	.702	.078
	.710	.070					.972	.166	.786	.166	.786	.160
	.976	.205							.858	.243	.864	.217
	1.072	.177							.919	.261	.912	.223
	1.110	.123							.967	.141	.985	.050
CN=		.3572		.3519		.3676		.3492		.3247		.2573
CM=		.0281		-.0377		-.0616		-.0783		-.0750		-.0642

$\alpha = 3.94^{\circ}$; $C_L = 0.386$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	-.163	-.021	-.725	.023	-1.033	.025	-.880	.022	-.764	.018	-.650
	-.567	-.308	.035	-.689	.068	-.652	.079	-.611	.075	-.614	.077	-.499
	-.452	-.351	.105	-.571	.209	-.440	.133	-.489	.129	-.433	.129	-.367
	-.311	-.360	.178	-.482	.294	-.382	.214	-.398	.201	-.367	.209	-.292
	-.023	-.305	.286	-.398	.404	-.303	.295	-.330	.294	-.320	.293	-.254
	.133	-.237	.396	-.328	.497	-.289	.407	-.302	.397	-.278	.494	-.206
	.272	-.191	.514	-.265	.599	-.260	.502	-.285	.495	-.265	.590	-.214
	.416	-.160	.618	-.228	.700	-.262	.601	-.273	.594	-.260	.693	-.227
	.565	-.116	.733	-.192	.864	-.221	.698	-.283	.693	-.268	.777	-.237
	.713	-.107	.835	-.164	.926	-.155	.863	-.261	.784	-.271	.861	-.219
	.854	-.107	.919	-.114			.923	-.188	.856	-.258	.918	-.193
	.980	-.100	.987	-.036			.977	-.067	.926	-.192	.972	-.073
	1.074	-.081							.977	-.059		
						LOWER SURFACE						
	-.660	.039	-.022	.337	.024	.330	.074	.097	.019	.264	.020	.254
	-.616	.044	.038	.123	.075	.091	.130	.150	.066	.044	.076	-.011
	-.462	-.007	.101	.045	.297	-.043	.258	-.016	.136	-.032	.136	-.049
	-.329	-.020	.185	-.019	.400	-.035	.397	-.035	.214	-.011	.221	-.058
	-.172	-.053	.398	-.043	.604	-.045	.501	-.034	.292	-.030	.295	-.060
	-.030	-.084	.737	.116	.785	.172	.603	.022	.403	-.025	.396	-.051
	.128	-.115			.967	.168	.784	.154	.489	-.045	.497	-.048
	.418	-.033			1.000	.006	.868	.229	.594	-.033	.597	-.020
	.564	-.002					.923	.279	.700	.079	.702	.078
	.710	.084					.972	.170	.786	.173	.786	.156
	.976	.208							.858	.249	.864	.219
	1.072	.179							.919	.262	.912	.221
	1.110	.128							.967	.143	.985	.050
CN=		.4028		.4054		.4132		.4106		.3767		.3022
CM=		.0341		-.0394		-.0617		-.0797		-.0766		-.0641

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = 4.96^{\circ}$; $C_L = 0.474$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.201	-.021	-1.106	.023	-1.350	.025	-1.110	.022	-1.102	.018	-.964
	-.567	-.373	.035	-.783	.068	-.849	.079	-.756	.075	-.764	.077	-.636
	-.452	-.377	.105	-.637	.209	-.499	.133	-.570	.129	-.544	.129	-.456
	-.311	-.406	.178	-.535	.294	-.413	.214	-.456	.201	-.431	.209	-.334
	-.023	-.330	.286	-.436	.404	-.352	.295	-.384	.294	-.353	.293	-.292
	.133	-.252	.396	-.359	.497	-.316	.407	-.329	.397	-.300	.494	-.227
	.272	-.215	.514	-.287	.599	-.289	.502	-.313	.495	-.291	.590	-.226
	.416	-.176	.618	-.246	.700	-.270	.601	-.291	.594	-.280	.693	-.237
	.565	-.139	.733	-.197	.864	-.219	.698	-.288	.693	-.283	.777	-.239
	.713	-.125	.835	-.163	.926	-.146	.863	-.257	.784	-.272	.861	-.220
	.854	-.128	.919	-.101			.923	-.185	.856	-.257	.918	-.189
	.980	-.115	.987	-.032			.977	-.059	.926	-.185	.972	-.074
	1.074	-.094							.977	-.051		
LOWER SURFACE												
	-.660	.033	-.022	.371	.024	.415	.074	.160	.019	.389	.020	.360
	-.616	.068	.038	.185	.075	.151	.130	.096	.066	.140	.076	.094
	-.462	.023	.101	.095	.297	.008	.298	.025	.136	.067	.136	.004
	-.329	-.001	.185	.044	.400	-.008	.397	.001	.214	.034	.221	-.010
	-.172	-.031	.398	-.024	.604	-.027	.501	-.012	.292	.022	.295	-.033
	-.030	-.066	.737	.120	.785	.176	.603	.042	.403	.009	.396	-.023
	.128	-.094			.967	.162	.784	.164	.489	-.018	.497	-.039
	.418	-.015			1.000	-.010	.868	.244	.594	-.016	.597	-.006
	.564	.022					.923	.281	.700	.084	.702	.081
	.710	.095					.972	.172	.786	.179	.786	.159
	.976	.221							.858	.256	.864	.217
	1.072	.194							.919	.257	.912	.218
	1.110	.135							.967	.138	.985	.042
CN=	.4834		.4788		.4942		.4813		.4595		.3711	
CM=	.0478		-.0326		-.0571		-.0786		-.0713		-.0585	

$\alpha = 5.99^{\circ}$; $C_L = 0.561$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.250	-.021	-1.441	.023	-1.733	.025	-1.554	.022	-1.495	.018	-1.282
	-.567	-.416	.035	-.957	.068	-1.061	.079	-.912	.075	-.905	.077	-.782
	-.452	-.425	.105	-.727	.209	-.576	.133	-.687	.129	-.653	.129	-.563
	-.311	-.436	.178	-.604	.294	-.469	.214	-.526	.201	-.505	.209	-.406
	-.023	-.356	.286	-.495	.404	-.387	.295	-.441	.294	-.409	.293	-.337
	.133	-.274	.396	-.396	.497	-.343	.407	-.370	.397	-.348	.494	-.253
	.272	-.229	.514	-.314	.599	-.303	.502	-.343	.495	-.330	.590	-.252
	.416	-.191	.618	-.268	.700	-.283	.601	-.314	.594	-.305	.693	-.249
	.565	-.157	.733	-.206	.864	-.211	.698	-.309	.693	-.298	.777	-.253
	.713	-.144	.835	-.156	.926	-.137	.863	-.257	.784	-.284	.861	-.227
	.854	-.141	.919	-.094			.923	-.181	.856	-.251	.918	-.190
	.980	-.132	.987	-.035			.977	-.064	.926	-.165	.972	-.076
	1.074	-.107							.977	-.054		
LOWER SURFACE												
	-.660	.028	-.022	.377	.024	.468	.074	.245	.019	.423	.020	.439
	-.616	.072	.038	.250	.075	.255	.130	.151	.066	.237	.076	.164
	-.462	.042	.101	.149	.297	.042	.298	.055	.136	.117	.136	.058
	-.329	.017	.185	.083	.400	.021	.397	.032	.214	.076	.221	.019
	-.172	-.017	.398	.008	.604	-.015	.501	.017	.292	.060	.295	.007
	-.030	-.033	.737	.133	.785	.183	.603	.058	.403	.035	.396	-.005
	.128	-.062			.967	.162	.784	.163	.489	-.013	.497	-.022
	.418	-.002			1.000	-.022	.868	.244	.594	-.000	.597	-.002
	.564	.035					.923	.283	.700	.092	.702	.080
	.710	.108					.972	.169	.786	.183	.786	.149
	.976	.233							.858	.252	.864	.205
	1.072	.189							.919	.266	.912	.218
	1.110	.139							.967	.134	.985	.029
CN=	.5570		.5647		.5845		.5713		.5401		.4422	
CM=	.0612		-.0290		-.0499		-.0744		-.0672		-.0536	

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = 6.91^{\circ}$; $C_L = 0.635$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.288	-.021	-1.688	.023	-2.171	.025	-2.001	.022	-1.914	.018	-1.622
-.567	-.470	.035	-1.063	.068	-1.201	.079	-1.066	.075	-1.055	.077	-.902
-.452	-.466	.105	-.798	.209	-.626	.133	-.767	.129	-.737	.129	-.631
-.311	-.467	.178	-.685	.294	-.515	.214	-.591	.201	-.578	.209	-.448
-.023	-.377	.286	-.530	.404	-.420	.295	-.484	.294	-.456	.293	-.374
.133	-.299	.396	-.425	.497	-.369	.407	-.403	.397	-.389	.494	-.271
.272	-.249	.514	-.328	.599	-.316	.502	-.362	.495	-.347	.590	-.266
.416	-.212	.618	-.280	.700	-.288	.601	-.327	.594	-.313	.693	-.257
.565	-.172	.733	-.213	.864	-.198	.698	-.312	.693	-.301	.777	-.258
.713	-.148	.835	-.154	.926	-.130	.863	-.246	.784	-.283	.861	-.218
.854	-.149	.919	-.089			.923	-.164	.856	-.237	.918	-.182
.980	-.138	.987	-.034			.977	-.061	.926	-.159	.972	-.072
1.074	-.113							.977	-.047		
				LOWER SURFACE							
-.660	.010	-.022	.373	.024	.499	.074	.295	.019	.496	.020	.473
-.616	.083	.038	.289	.075	.316	.130	.198	.066	.265	.076	.223
-.462	.070	.101	.197	.297	.056	.238	.071	.136	.171	.136	.114
-.329	.042	.185	.120	.400	.050	.397	.058	.214	.111	.221	.057
-.172	.012	.398	.044	.604	.003	.501	.031	.292	.088	.295	.025
-.030	-.019	.737	.142	.735	.187	.603	.072	.403	.043	.396	.009
.128	-.045			.967	.156	.784	.163	.489	.018	.497	-.013
.418	.023			1.000	-.033	.868	.248	.594	.011	.597	-.004
.564	.054					.923	.290	.700	.100	.702	.082
.710	.134					.972	.164	.786	.175	.786	.144
.976	.241							.858	.254	.864	.205
1.072	.199							.919	.260	.912	.205
1.110	.142							.967	.132	.985	.014
CN=	.6310		.6328		.6513		.6416		.6077		.4965
CM=	.0761		-.0265		-.0442		-.0675		-.0599		-.0468

$\alpha = 7.99^{\circ}$; $C_L = 0.723$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.598	-.021	-2.081	.023	-2.598	.025	-2.126	.022	-2.126	.018	-1.411
-.557	-.511	.035	-1.257	.068	-1.314	.079	-1.194	.075	-1.374	.077	-1.062
-.452	-.522	.105	-.914	.209	-.679	.133	-.854	.129	-.937	.129	-.743
-.311	-.504	.178	-.753	.294	-.536	.214	-.653	.201	-.825	.209	-.533
-.023	-.416	.285	-.572	.404	-.446	.295	-.522	.294	-.517	.293	-.421
.133	-.327	.395	-.452	.497	-.387	.407	-.438	.397	-.411	.494	-.289
.272	-.279	.514	-.353	.599	-.316	.502	-.373	.495	-.379	.590	-.271
.416	-.229	.618	-.294	.700	-.289	.601	-.331	.594	-.324	.693	-.242
.555	-.188	.733	-.220	.864	-.188	.698	-.313	.693	-.292	.777	-.234
.713	-.164	.835	-.158	.926	-.117	.863	-.227	.784	-.271	.861	-.206
.854	-.165	.919	-.094			.923	-.145	.856	-.229	.918	-.155
.980	-.143	.987	-.053			.977	-.071	.926	-.143	.972	-.093
1.074	-.113							.977	-.061		
				LOWER SURFACE							
-.660	-.011	-.022	.342	.024	.525	.074	.351	.019	.517	.020	.510
-.616	.091	.038	.339	.075	.354	.130	.235	.066	.345	.076	.284
-.462	.081	.101	.245	.297	.114	.298	.119	.136	.230	.136	.174
-.329	.069	.185	.165	.400	.066	.397	.089	.214	.166	.221	.089
-.172	.036	.398	.072	.604	.017	.501	.058	.292	.122	.295	.063
-.030	.006	.737	.148	.785	.193	.603	.097	.403	.083	.396	.030
.128	-.022			.967	.157	.784	.173	.489	.049	.497	.006
.418	.047			1.000	-.035	.868	.256	.594	.034	.597	.015
.564	.077					.923	.297	.700	.110	.702	.080
.710	.152					.972	.168	.786	.186	.786	.145
.976	.250							.858	.258	.864	.203
1.072	.218							.919	.262	.912	.198
1.110	.149							.967	.126	.985	.000
CN=	.7144		.7168		.7140		.7023		.7068		.5349
CM=	.0929		-.0220		-.0383		-.0659		-.0542		-.0442

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = 8.94^{\circ}$; $C_L = 0.801$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.427	-.021	-2.736	.023	-1.591	.025	-2.167	.022	-1.415	.018	-.699
-.567	-.573	.035	-1.405	.068	-1.587	.079	-1.194	.075	-1.282	.077	-.712
-.452	-.548	.105	-1.002	.209	-1.363	.133	-.856	.129	-1.192	.129	-.651
-.311	-.533	.178	-.810	.294	-.893	.214	-.661	.201	-1.165	.209	-.622
-.023	-.432	.286	-.636	.404	-.582	.295	-.548	.294	-.916	.293	-.606
.133	-.343	.396	-.507	.497	-.384	.407	-.442	.397	-.635	.494	-.442
.272	-.294	.514	-.390	.599	-.349	.502	-.379	.495	-.558	.590	-.390
.416	-.250	.618	-.317	.700	-.238	.601	-.324	.594	-.288	.693	-.316
.565	-.205	.733	-.226	.864	-.187	.698	-.283	.693	-.262	.777	-.268
.713	-.178	.835	-.172	.926	-.144	.863	-.223	.784	-.236	.861	-.215
.854	-.174	.919	-.115			.923	-.163	.856	-.213	.918	-.197
.980	-.161	.987	-.072			.977	-.190	.926	-.162	.972	-.173
1.074	-.120							.977	-.075		
				LOWER SURFACE							
-.660	-.031	-.022	.320	.024	.542	.074	.380	.019	.529	.020	.505
-.616	.096	.038	.374	.075	.407	.130	.272	.066	.374	.076	.279
-.462	.120	.101	.288	.297	.145	.298	.140	.136	.247	.136	.167
-.329	.079	.185	.222	.400	.098	.397	.101	.214	.175	.221	.092
-.172	.063	.398	.110	.604	.025	.501	.071	.292	.132	.295	.060
-.030	.028	.737	.161	.785	.203	.603	.092	.403	.096	.396	.033
.128	.013			.967	.178	.784	.152	.489	.053	.497	.005
.418	.074			1.000	-.001	.868	.239	.594	.035	.597	.001
.564	.096					.923	.280	.700	.114	.702	.071
.710	.164					.972	.120	.786	.188	.786	.119
.976	.266							.858	.261	.864	.172
1.072	.227							.919	.267	.912	.176
1.110	.157							.967	.136	.985	-.080
CN=	.7910		.8247		.8430		.7130		.7908		.5481
CM=	.1046		-.0195		-.0477		-.0634		-.0663		-.0694

$\alpha = 9.93^{\circ}$; $C_L = 0.875$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.506	-.021	-3.006	.023	-1.336	.025	-2.160	.022	-1.146	.018	-.640
-.567	-.650	.035	-1.656	.068	-1.347	.079	-1.144	.075	-1.103	.077	-.588
-.452	-.599	.105	-1.144	.209	-1.227	.133	-.823	.129	-1.103	.129	-.591
-.311	-.576	.178	-.897	.294	-1.152	.214	-.633	.201	-1.089	.209	-.529
-.023	-.460	.286	-.690	.404	-.965	.295	-.557	.294	-1.058	.293	-.532
.133	-.364	.396	-.524	.497	-.857	.407	-.474	.397	-.950	.494	-.423
.272	-.316	.514	-.409	.599	-.623	.502	-.395	.495	-.761	.590	-.388
.416	-.271	.618	-.321	.700	-.303	.601	-.348	.594	-.588	.693	-.348
.565	-.222	.733	-.245	.864	-.257	.698	-.309	.693	-.412	.777	-.317
.713	-.210	.835	-.177	.926	-.261	.863	-.266	.784	-.286	.861	-.284
.854	-.202	.919	-.124			.923	-.258	.856	-.267	.918	-.257
.980	-.177	.987	-.086			.977	-.247	.926	-.188	.972	-.244
1.074	-.125							.977	-.123		
				LOWER SURFACE							
-.660	-.049	-.022	.253	.024	.551	.074	.389	.019	.528	.020	.494
-.616	.092	.038	.387	.075	.430	.130	.282	.066	.382	.076	.283
-.462	.138	.101	.325	.297	.164	.298	.149	.136	.264	.136	.165
-.329	.102	.185	.238	.430	.109	.397	.094	.214	.182	.221	.092
-.172	.082	.398	.132	.604	.032	.501	.066	.292	.141	.295	.060
-.030	.051	.737	.166	.785	.193	.603	.086	.403	.097	.396	.017
.128	.037			.967	.122	.784	.135	.489	.048	.497	-.013
.418	.096			1.000	-.045	.868	.221	.594	.037	.597	-.024
.564	.114					.923	.257	.700	.113	.702	.034
.710	.182					.972	.091	.786	.185	.786	.096
.976	.277							.858	.255	.864	.150
1.072	.222							.919	.260	.912	.141
1.110	.153							.967	.125	.985	-.130
CN=	.8773		.8999		.9614		.7253		.8792		.5144
CM=	.1208		-.0156		-.0944		-.0718		-.1047		-.0741

[redacted]

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Concluded.

$\alpha = 10.92^{\circ}$; $C_L = 0.942$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.562	-.021	-3.038	.023	-1.191	.025	-2.236	.022	-1.161	.018	-.591
-.567	-.721	.035	-2.415	.068	-1.138	.079	-1.096	.075	-1.150	.077	-.589
-.452	-.676	.105	-1.629	.209	-1.025	.133	-.738	.129	-1.138	.129	-.583
-.311	-.631	.178	-.910	.294	-.944	.214	-.595	.201	-1.147	.209	-.540
-.023	-.499	.285	-.686	.404	-.934	.295	-.540	.294	-1.144	.293	-.510
.133	-.418	.396	-.532	.497	-.864	.407	-.463	.397	-1.092	.494	-.426
.272	-.355	.514	-.386	.599	-.780	.502	-.423	.495	-.954	.590	-.383
.416	-.315	.618	-.315	.700	-.714	.601	-.373	.594	-.718	.693	-.360
.565	-.276	.732	-.245	.864	-.515	.698	-.340	.693	-.490	.777	-.334
.713	-.235	.835	-.173	.926	-.444	.863	-.318	.784	-.341	.861	-.321
.854	-.235	.919	-.123			.923	-.309	.856	-.238	.918	-.305
.980	-.197	.967	-.074			.977	-.280	.925	-.186	.972	-.303
1.074	-.145							.977	-.107		
				LOWER SURFACE							
-.650	-.055	-.022	.209	.024	.567	.074	.405	.019	.538	.020	.501
-.616	.100	.039	.434	.075	.462	.130	.313	.066	.419	.076	.304
-.462	.165	.101	.357	.297	.193	.298	.150	.136	.284	.136	.185
-.329	.146	.185	.279	.400	.132	.397	.102	.214	.198	.221	.103
-.172	.121	.398	.158	.604	.035	.501	.063	.292	.155	.295	.067
-.030	.106	.737	.199	.785	.176	.603	.078	.403	.105	.396	.024
.128	.065			.957	.060	.784	.122	.489	.066	.497	-.018
.418	.127			1.000	-.292	.868	.220	.594	.037	.597	-.024
.564	.145					.923	.249	.700	.112	.702	.028
.716	.209					.972	.063	.786	.190	.786	.085
.976	.293							.858	.259	.864	.140
1.072	.245							.919	.267	.912	.140
1.110	.174							.967	.131	.985	-.156
CN=	1.0128		1.0077		1.0127		.7351		.9630		.5217
CM=	.1336		-.0040		-.1533		-.0786		-.1195		-.0788

[redacted]

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80.

$\alpha = -5.07^{\circ}$; $C_L = -0.481$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.037	-.021	.345	.023	.395	.025	.368	.022	.467	.018	.464
-.567	.005	.035	.197	.068	.249	.079	.261	.075	.265	.077	.221
-.452	-.086	.105	.125	.134	.154	.133	.162	.129	.196	.129	.160
-.311	-.131	.178	.055	.209	.094	.214	.105	.201	.134	.209	.099
-.023	-.096	.286	.013	.294	.060	.295	.073	.294	.090	.293	.032
.133	-.025	.396	.001	.404	.024	.407	.027	.397	.045	.494	-.070
.272	.006	.514	-.012	.497	-.018	.502	-.019	.495	-.013	.590	-.133
.416	.028	.618	-.031	.599	-.052	.601	-.065	.594	-.058	.693	-.217
.565	.050	.733	-.047	.700	-.100	.698	-.132	.693	-.130	.777	-.291
.713	.043	.835	-.084	.864	-.171	.863	-.239	.784	-.175	.861	-.371
.854	.009	.919	-.076	.926	-.152	.923	-.232	.856	-.207	.918	-.453
.980	-.022	.987	-.019	.975	-.048	.977	-.122	.926	-.167	.972	-.426
1.074	-.043							.977	-.061		
1.122	-.021										
					LOWER SURFACE						
-.660	-.062	-.022	-1.045	.024	-1.505	.025	-1.614	.019	-.772	.020	-.410
-.616	-.176	.038	-1.103	.075	-1.448	.074	-1.553	.066	-.765	.076	-.361
-.572	-.209	.101	-.880	.297	-.626	.130	-1.381	.136	-.730	.136	-.326
-.462	-.227	.185	-.710	.400	-.368	.298	-.366	.214	-.719	.221	-.311
-.329	-.234	.398	-.406	.604	-.235	.397	-.240	.292	-.714	.295	-.304
-.172	-.265	.737	-.015	.785	.051	.501	-.190	.403	-.688	.396	-.257
-.030	-.327			.967	.117	.603	-.086	.489	-.668	.497	-.232
.128	-.401			1.000	.055	.703	-.019	.594	-.617	.597	-.205
.418	-.363					.784	.028	.700	-.567	.702	-.209
.564	-.277					.868	.063	.786	-.442	.786	-.221
.710	-.114					.923	.076	.858	-.316	.864	-.214
.976	.136					.972	.070	.919	-.269	.912	-.207
1.110	.126							.967	-.096	.985	-.207
CN=	-.3235		-.4645		-.4570		-.3956		-.5946		-.1747
CM=	-.0657		-.0486		-.0699		-.0932		.0546		-.0379

$\alpha = -4.10^{\circ}$; $C_L = -0.399$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.028	-.021	.302	.023	.336	.025	.307	.022	.432	.018	.448
-.567	-.020	.035	.132	.068	.189	.079	.202	.075	.229	.077	.192
-.452	-.119	.105	.055	.134	.090	.133	.122	.129	.163	.129	.128
-.311	-.153	.178	.016	.209	.035	.214	.076	.201	.105	.209	.067
-.023	-.121	.286	-.025	.294	.020	.295	.041	.294	.062	.293	.012
.133	-.054	.396	-.037	.404	-.013	.407	.004	.397	.025	.494	-.079
.272	-.027	.514	-.042	.497	-.048	.502	-.041	.495	-.029	.590	-.127
.416	.008	.618	-.059	.599	-.066	.601	-.077	.594	-.073	.693	-.196
.565	.027	.733	-.071	.700	-.115	.698	-.130	.693	-.140	.777	-.262
.713	.021	.835	-.094	.864	-.164	.863	-.213	.784	-.177	.861	-.328
.854	-.009	.919	-.085	.926	-.136	.923	-.193	.856	-.206	.918	-.404
.980	-.033	.987	-.021	.975	-.041	.977	-.064	.926	-.170	.972	-.398
1.074	-.051							.977	-.008		
1.122	-.021										
					LOWER SURFACE						
-.660	-.046	-.022	-.772	.024	-1.450	.025	-1.541	.019	-.826	.020	-.523
-.616	-.136	.038	-.844	.075	-1.347	.074	-1.527	.066	-.802	.076	-.411
-.572	-.170	.101	-.791	.297	-.432	.130	-1.372	.136	-.783	.136	-.394
-.462	-.202	.185	-.631	.400	-.355	.298	-.326	.214	-.755	.221	-.349
-.329	-.211	.398	-.378	.604	-.219	.397	-.271	.292	-.738	.295	-.337
-.172	-.244	.737	.005	.785	.089	.501	-.192	.403	-.677	.396	-.272
-.030	-.303			.967	.166	.603	-.046	.489	-.625	.497	-.236
.128	-.363			1.000	.059	.703	.032	.594	-.472	.597	-.213
.418	-.325					.784	.046	.700	-.314	.702	-.198
.564	-.237					.868	.082	.786	-.171	.786	-.214
.710	-.093					.923	.107	.858	-.100	.864	-.198
.976	.144					.972	.095	.919	-.012	.912	-.198
1.110	.130							.967	.107	.985	-.201
CN=	-.2359		-.3629		-.3634		-.3614		-.4872		-.1942
CM=	-.0565		-.0479		-.0774		-.0932		-.0009		-.0340

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{o}$

(c) M = 0.80. Continued.

$\alpha = -3.16^{o}$; $C_L = -0.307$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.033	-.021	.264	.023	.248	.025	.258	.022	.387	.018	.420
-.567	-.033	.035	.062	.068	.103	.079	.141	.075	.160	.077	.165
-.452	-.138	.105	-.003	.134	.041	.133	.073	.129	.114	.129	.104
-.311	-.179	.178	-.037	.209	.006	.214	.022	.201	.064	.209	.045
-.023	-.147	.286	-.069	.294	-.017	.295	.006	.294	.026	.293	-.008
.133	-.083	.396	-.084	.404	-.049	.407	-.032	.397	.001	.494	-.075
.272	-.047	.514	-.070	.497	-.074	.502	-.068	.495	-.053	.590	-.113
.416	-.012	.618	-.080	.599	-.103	.601	-.103	.594	-.097	.693	-.157
.565	.008	.733	-.089	.700	-.134	.698	-.151	.693	-.149	.777	-.207
.713	.007	.835	-.103	.864	-.173	.863	-.220	.784	-.189	.861	-.189
.854	-.019	.919	-.095	.926	-.129	.923	-.174	.856	-.223	.918	-.246
.980	-.044	.987	-.027	.975	-.040	.977	-.041	.926	-.173	.972	-.182
1.074	-.054							.977	-.038		
1.122	-.029										
				LOWER SURFACE							
-.660	-.029	-.022	-.515	.024	-1.285	.025	-1.450	.019	-1.013	.020	-.856
-.616	-.130	.038	-.577	.075	-1.158	.074	-1.404	.066	-.997	.076	-.791
-.572	-.162	.101	-.625	.297	-.413	.130	-1.215	.136	-.833	.136	-.494
-.462	-.181	.185	-.546	.400	-.340	.298	-.360	.214	-.749	.221	-.504
-.329	-.155	.398	-.345	.604	-.197	.397	-.302	.292	-.676	.295	-.393
-.172	-.223	.737	.025	.785	.083	.501	-.224	.403	-.537	.396	-.353
-.030	-.285			.967	.182	.603	-.058	.489	-.349	.497	-.247
.128	-.345			1.000	.046	.703	.052	.594	-.242	.597	-.194
.418	-.283					.784	.085	.700	-.089	.702	-.117
.564	-.220					.868	.119	.786	-.001	.786	-.003
.710	-.069					.923	.154	.858	.084	.864	-.073
.976	.149					.972	.118	.919	.130	.912	-.075
1.110	.127							.967	.102	.985	-.007
CN=	-.1688		-.2526		-.2890		-.3064		-.3573		-.2453
CM=	-.0514		-.0462		-.0757		-.0945		-.0546		-.0456

$\alpha = -2.18^{o}$; $C_L = -0.208$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.014	-.021	.191	.023	.185	.025	.166	.022	.333	.018	.388
-.567	-.057	.035	-.009	.068	.034	.079	.057	.075	.110	.077	.128
-.452	-.162	.105	-.067	.134	-.028	.133	.023	.129	.070	.129	.064
-.311	-.222	.178	-.096	.209	-.055	.214	-.020	.201	.019	.209	.019
-.023	-.168	.286	-.123	.294	-.067	.295	-.039	.294	-.000	.293	-.026
.133	-.094	.396	-.111	.404	-.079	.407	-.059	.397	-.038	.494	-.086
.272	-.073	.514	-.100	.497	-.096	.502	-.085	.495	-.075	.590	-.114
.416	-.029	.618	-.113	.599	-.121	.601	-.124	.594	-.104	.693	-.150
.565	-.014	.733	-.108	.700	-.159	.698	-.164	.693	-.165	.777	-.178
.713	-.008	.835	-.119	.864	-.182	.863	-.218	.784	-.200	.861	-.164
.854	-.032	.919	-.107	.926	-.133	.923	-.168	.856	-.224	.918	-.143
.980	-.051	.987	-.031	.975	-.041	.977	-.032	.926	-.179	.972	-.030
1.074	-.054							.977	-.027		
1.122	-.023										
				LOWER SURFACE							
-.660	-.003	-.022	-.331	.024	-.938	.025	-1.266	.019	-1.138	.020	-1.294
-.616	-.091	.038	-.495	.075	-.795	.074	-1.253	.066	-1.051	.076	-1.204
-.572	-.141	.101	-.501	.297	-.367	.130	-.601	.136	-.673	.136	-.618
-.462	-.149	.185	-.456	.400	-.309	.298	-.351	.214	-.487	.221	-.392
-.329	-.169	.398	-.309	.604	-.172	.397	-.298	.292	-.367	.295	-.289
-.172	-.204	.737	.045	.785	.101	.501	-.225	.403	-.287	.396	-.202
-.030	-.251			.967	.196	.603	-.066	.489	-.233	.497	-.136
.128	-.309			1.000	.042	.703	.056	.594	-.149	.597	-.053
.418	-.243					.784	.109	.700	-.012	.702	.046
.564	-.174					.868	.169	.786	.087	.786	.092
.710	-.041					.923	.185	.858	.157	.864	.134
.976	.163					.972	.149	.919	.165	.912	.140
1.110	.131							.967	.118	.985	.081
CN=	-.0805		-.1542		-.1688		-.1836		-.2133		-.1932
CM=	-.0400		-.0486		-.0714		-.0877		-.0781		-.0837

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Continued.

$\alpha = -1.17^{\circ}$; $C_L = -0.102$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.007	-.021	.110	.023	.043	.025	.064	.022	.231	.018	.322
-.567	-.092	.035	-.100	.068	-.107	.079	-.034	.075	.016	.077	.073
-.452	-.205	.105	-.155	.134	-.109	.133	-.055	.129	-.004	.129	.011
-.311	-.243	.178	-.160	.209	-.121	.214	-.073	.201	-.040	.209	-.034
-.023	-.210	.286	-.174	.294	-.116	.295	-.091	.294	-.055	.293	-.063
.133	-.124	.396	-.167	.404	-.132	.407	-.111	.397	-.078	.494	-.108
.272	-.097	.514	-.134	.497	-.149	.502	-.132	.495	-.112	.590	-.138
.416	-.066	.618	-.148	.599	-.155	.601	-.161	.594	-.148	.693	-.168
.565	-.034	.733	-.131	.700	-.181	.698	-.196	.693	-.187	.777	-.184
.713	-.036	.835	-.134	.864	-.198	.863	-.239	.784	-.227	.861	-.167
.854	-.050	.919	-.113	.926	-.141	.923	-.171	.856	-.242	.918	-.140
.980	-.063	.987	-.027	.975	-.045	.977	-.028	.926	-.181	.972	-.011
1.074	-.063							.977	-.026		
1.122	-.028										
					LOWER SURFACE						
-.660	.004	-.022	-.197	.024	-.613	.025	-.980	.019	-1.116	.020	-1.110
-.616	-.062	.038	-.356	.075	-.613	.074	-.813	.066	-.971	.076	-.864
-.572	-.102	.101	-.415	.297	-.315	.130	-.451	.136	-.450	.136	-.582
-.462	-.137	.185	-.365	.400	-.275	.298	-.312	.214	-.373	.221	-.347
-.329	-.153	.398	-.277	.604	-.154	.397	-.262	.292	-.294	.295	-.263
-.172	-.198	.737	.056	.785	.122	.501	-.211	.403	-.251	.396	-.202
-.030	-.242			.967	.200	.603	-.064	.489	-.220	.497	-.134
.128	-.279			1.000	.030	.703	.059	.594	-.152	.597	-.044
.418	-.207					.784	.114	.700	.015	.702	.077
.564	-.145					.868	.185	.786	.114	.786	.138
.710	-.026					.923	.191	.858	.167	.864	.165
.976	.172					.972	.148	.919	.196	.912	.168
1.110	.126							.967	.152	.985	.094
CN=	.0016		-.0556		-.0574		-.0720		-.1190		-.1178
CM=	-.0314		-.0487		-.0712		-.0849		-.0846		-.0847

$\alpha = -0.01^{\circ}$; $C_L = 0.017$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.016	-.021	-.019	.023	-.181	.025	-.083	.022	.092	.018	.221
-.567	-.135	.035	-.221	.068	-.233	.079	-.180	.075	-.091	.077	-.043
-.452	-.239	.105	-.240	.134	-.208	.133	-.141	.129	-.089	.129	-.075
-.311	-.289	.178	-.259	.209	-.193	.214	-.154	.201	-.108	.209	-.087
-.023	-.231	.286	-.220	.294	-.184	.295	-.140	.294	-.119	.293	-.110
.133	-.158	.396	-.210	.404	-.178	.407	-.164	.397	-.124	.494	-.132
.272	-.115	.514	-.169	.497	-.177	.502	-.166	.495	-.145	.590	-.152
.416	-.089	.618	-.174	.599	-.197	.601	-.184	.594	-.183	.693	-.187
.565	-.059	.733	-.151	.700	-.213	.698	-.230	.693	-.220	.777	-.201
.713	-.051	.835	-.152	.864	-.214	.863	-.246	.784	-.251	.861	-.176
.854	-.063	.919	-.120	.926	-.150	.923	-.172	.856	-.255	.918	-.145
.980	-.083	.987	-.026	.975	-.053	.977	-.025	.926	-.184	.972	-.011
1.074	-.075							.977	-.028		
1.122	-.024										
					LOWER SURFACE						
-.660	.016	-.022	-.019	.024	-.286	.025	-.468	.019	-.710	.020	-.621
-.616	-.037	.038	-.227	.075	-.424	.074	-.457	.066	-.593	.076	-.585
-.572	-.069	.101	-.285	.297	-.251	.130	-.341	.136	-.412	.136	-.526
-.462	-.114	.185	-.293	.400	-.216	.298	-.247	.214	-.289	.221	-.268
-.329	-.131	.398	-.229	.604	-.126	.397	-.216	.292	-.237	.295	-.221
-.172	-.193	.737	.077	.785	.148	.501	-.180	.403	-.198	.396	-.179
-.030	-.213			.967	.197	.603	-.053	.489	-.193	.497	-.128
.128	-.253			1.000	.024	.703	.067	.594	-.141	.597	-.049
.418	-.168					.784	.130	.700	.033	.702	.086
.564	-.120					.868	.187	.786	.141	.786	.164
.710	.000					.923	.205	.858	.198	.864	.206
.976	.184					.972	.157	.919	.226	.912	.210
1.110	.137							.967	.162	.985	.091
CN=	.0960		.0568		.0683		.0542		-.0004		-.0169
CM=	-.0179		-.0493		-.0714		-.0798		-.0853		-.0817

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Continued.

$\alpha = 0.96^{\circ}$; $C_L = 0.115$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
					UPPER SURFACE						
-.660	-.038	-.021	-.186	.023	-.331	.025	-.252	.022	-.085	.018	.043
-.567	-.169	.035	-.327	.068	-.370	.079	-.308	.075	-.214	.077	-.156
-.452	-.269	.105	-.323	.134	-.295	.133	-.253	.129	-.189	.129	-.149
-.311	-.316	.178	-.303	.209	-.269	.214	-.222	.201	-.179	.209	-.144
-.023	-.274	.286	-.288	.294	-.238	.295	-.201	.294	-.174	.293	-.154
.133	-.182	.396	-.255	.404	-.225	.407	-.203	.397	-.160	.494	-.163
.272	-.140	.514	-.208	.497	-.220	.502	-.207	.495	-.195	.590	-.181
.416	-.113	.618	-.191	.599	-.226	.601	-.222	.594	-.206	.693	-.207
.565	-.074	.733	-.169	.700	-.243	.698	-.255	.693	-.250	.777	-.228
.713	-.062	.835	-.165	.864	-.227	.863	-.251	.784	-.262	.861	-.198
.854	-.091	.919	-.115	.926	-.157	.923	-.178	.856	-.266	.918	-.161
.980	-.100	.987	-.030	.975	-.057	.977	-.032	.926	-.191	.972	-.029
1.074	-.083							.977	-.028		
1.122	-.045										
					LOWER SURFACE						
-.660	.030	-.022	.109	.024	-.129	.025	-.247	.019	-.428	.020	-.369
-.616	-.025	.038	-.138	.075	-.257	.074	-.279	.066	-.389	.076	-.458
-.572	-.057	.101	-.202	.297	-.215	.130	-.278	.136	-.317	.136	-.400
-.462	-.093	.185	-.212	.400	-.176	.298	-.190	.214	-.237	.221	-.253
-.329	-.106	.398	-.183	.604	-.114	.397	-.168	.292	-.180	.295	-.171
-.172	-.143	.737	.089	.785	.155	.501	-.155	.403	-.167	.396	-.158
-.030	-.185			.967	.198	.603	-.039	.489	-.162	.497	-.119
.128	-.229			1.000	.020	.703	.069	.594	-.119	.597	-.045
.418	-.142					.784	.153	.700	.049	.702	.091
.564	-.093					.868	.208	.786	.163	.786	.174
.710	.021					.923	.231	.858	.231	.864	.223
.976	.202					.972	.175	.919	.247	.912	.235
1.110	.130							.967	.165	.985	.079
CN=	.1778		.1551		.1637		.1546		.1034		.0698
CM=	-.0105		-.0487		-.0709		-.0819		-.0865		-.0822

$\alpha = 1.92^{\circ}$; $C_L = 0.213$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
					UPPER SURFACE						
-.660	-.060	-.021	-.316	.023	-.538	.025	-.437	.022	-.277	.018	-.125
-.567	-.185	.035	-.464	.068	-.553	.079	-.436	.075	-.356	.077	-.268
-.452	-.297	.105	-.423	.134	-.383	.133	-.327	.129	-.275	.129	-.236
-.311	-.346	.178	-.390	.209	-.340	.214	-.304	.201	-.257	.209	-.209
-.023	-.290	.286	-.355	.294	-.287	.295	-.250	.294	-.228	.293	-.189
.133	-.208	.396	-.290	.404	-.268	.407	-.235	.397	-.212	.494	-.183
.272	-.174	.514	-.238	.497	-.253	.502	-.239	.495	-.233	.590	-.194
.416	-.138	.618	-.217	.599	-.244	.601	-.253	.594	-.236	.693	-.216
.565	-.101	.733	-.188	.700	-.257	.698	-.285	.693	-.268	.777	-.236
.713	-.085	.835	-.174	.864	-.234	.863	-.270	.784	-.283	.861	-.205
.854	-.106	.919	-.115	.926	-.155	.923	-.183	.856	-.277	.918	-.171
.980	-.103	.987	-.023	.975	-.054	.977	-.037	.926	-.190	.972	-.034
1.074	-.092							.977	-.031		
1.122	-.047										
					LOWER SURFACE						
-.660	.032	-.022	.197	.024	.048	.025	-.082	.019	-.100	.020	-.103
-.616	.007	.038	-.040	.075	-.144	.074	-.157	.066	-.225	.076	-.290
-.572	-.014	.101	-.120	.297	-.152	.130	-.179	.136	-.213	.136	-.295
-.462	-.052	.185	-.150	.400	-.140	.298	-.147	.214	-.168	.221	-.224
-.329	-.092	.398	-.139	.604	-.101	.397	-.128	.292	-.157	.295	-.167
-.172	-.116	.737	.102	.785	.165	.501	-.122	.403	-.120	.396	-.131
-.030	-.162			.967	.194	.603	-.022	.489	-.133	.497	-.100
.128	-.194			1.000	.018	.703	.063	.594	-.100	.597	-.035
.418	-.120					.784	.153	.700	.060	.702	.092
.564	-.068					.868	.224	.786	.175	.786	.183
.710	.042					.923	.260	.858	.248	.864	.234
.976	.214					.972	.181	.919	.269	.912	.242
1.110	.140							.967	.162	.985	.069
CN=	.2603		.2513		.2551		.2467		.2059		.1470
CM=	.0000		-.0479		-.0678		-.0835		-.0856		-.0778

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Continued.

$\alpha = 2.40^{\circ}$; $C_L = 0.260$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.089	-.021	-.433	.023	-.727	.025	-.582	.022	-.399	.018	-.238
-.567	-.202	.035	-.514	.068	-.621	.079	-.502	.075	-.431	.077	-.358
-.452	-.329	.105	-.488	.134	-.432	.133	-.390	.129	-.349	.129	-.273
-.311	-.358	.178	-.410	.209	-.368	.214	-.340	.201	-.299	.209	-.243
-.023	-.297	.286	-.379	.294	-.325	.295	-.274	.294	-.263	.293	-.221
.133	-.220	.396	-.307	.404	-.275	.407	-.261	.397	-.231	.494	-.198
.272	-.186	.514	-.245	.497	-.266	.502	-.274	.495	-.248	.590	-.206
.416	-.151	.618	-.230	.599	-.254	.601	-.267	.594	-.253	.693	-.225
.565	-.100	.733	-.198	.700	-.263	.698	-.295	.693	-.280	.777	-.244
.713	-.088	.835	-.164	.864	-.232	.863	-.269	.784	-.293	.861	-.214
.854	-.118	.919	-.117	.926	-.152	.923	-.186	.856	-.290	.918	-.173
.980	-.112	.987	-.025	.975	-.046	.977	-.042	.926	-.182	.972	-.038
1.074	-.098							.977	-.034		
1.122	-.050										
					LOWER SURFACE						
-.660	.041	-.022	.249	.024	.127	.025	.040	.019	.008	.020	-.038
-.616	.007	.038	.010	.075	-.101	.074	-.124	.066	-.173	.076	-.231
-.572	-.008	.101	-.087	.297	-.132	.130	-.128	.136	-.163	.136	-.223
-.462	-.048	.185	-.113	.400	-.116	.298	-.118	.214	-.138	.221	-.187
-.329	-.075	.398	-.121	.604	-.092	.397	-.122	.292	-.136	.295	-.163
-.172	-.111	.737	.106	.785	.169	.501	-.116	.403	-.103	.396	-.117
-.030	-.144			.967	.193	.603	-.013	.489	-.113	.497	-.102
.128	-.178			1.000	.007	.703	.063	.594	-.094	.597	-.034
.418	-.103					.784	.157	.700	.063	.702	.093
.564	-.052					.868	.228	.786	.179	.786	.181
.710	.049					.923	.270	.858	.250	.864	.233
.976	.216					.972	.184	.919	.273	.912	.241
1.110	.139							.967	.156	.985	.066
CN=	.2999		.2958		.2990		.2970		.2556		.1889
CM=	.0069		-.0457		-.0651		-.0824		-.0847		-.0757

$\alpha = 2.86^{\circ}$; $C_L = 0.306$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.114	-.021	-.467	.023	-.914	.025	-.712	.022	-.549	.018	-.370
-.567	-.234	.035	-.537	.068	-.747	.079	-.608	.075	-.507	.077	-.453
-.452	-.324	.105	-.514	.134	-.472	.133	-.424	.129	-.383	.129	-.336
-.311	-.380	.178	-.450	.209	-.412	.214	-.373	.201	-.330	.209	-.278
-.023	-.318	.286	-.405	.294	-.352	.295	-.312	.294	-.290	.293	-.250
.133	-.236	.396	-.340	.404	-.305	.407	-.287	.397	-.252	.494	-.208
.272	-.193	.514	-.257	.497	-.286	.502	-.277	.495	-.267	.590	-.220
.416	-.162	.618	-.236	.599	-.270	.601	-.277	.594	-.272	.693	-.236
.565	-.132	.733	-.204	.700	-.274	.698	-.301	.693	-.288	.777	-.250
.713	-.111	.835	-.169	.864	-.234	.863	-.282	.784	-.294	.861	-.218
.854	-.114	.919	-.116	.926	-.155	.923	-.184	.856	-.287	.918	-.178
.980	-.111	.987	-.022	.975	-.056	.977	-.040	.926	-.183	.972	-.043
1.074	-.105							.977	-.036		
1.122	-.059										
					LOWER SURFACE						
-.660	.037	-.022	.264	.024	.199	.025	.112	.019	.085	.020	.042
-.616	.015	.038	.032	.075	-.056	.074	-.071	.066	-.107	.076	-.183
-.572	.004	.101	-.043	.297	-.111	.130	-.080	.136	-.136	.136	-.196
-.462	-.045	.185	-.088	.400	-.105	.298	-.101	.214	-.132	.221	-.162
-.329	-.067	.398	-.106	.604	-.080	.397	-.110	.292	-.110	.295	-.147
-.172	-.106	.737	.119	.785	.177	.501	-.099	.403	-.079	.396	-.100
-.030	-.134			.967	.193	.603	-.002	.489	-.098	.497	-.084
.128	-.167			1.000	.003	.703	.068	.594	-.083	.597	-.036
.418	-.092					.784	.151	.700	.070	.702	.094
.564	-.048					.868	.232	.786	.180	.786	.178
.710	.060					.923	.281	.858	.255	.864	.238
.976	.225					.972	.188	.919	.271	.912	.237
1.110	.144							.967	.157	.985	.064
CN=	.3408		.3342		.3509		.3434		.2990		.2318
CM=	.0104		-.0482		-.0646		-.0815		-.0839		-.0738

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Continued.

$\alpha = 3.34^{\circ}$; $C_L = 0.354$

	STA .133		STA .307		STA .480		STA .653		STA .804		STA .933	
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.123	-.021	-.590	.023	-1.083	.025	-.906	.022	-.667	.018	-.508
	-.567	-.244	.035	-.609	.068	-.899	.079	-.744	.075	-.624	.077	-.534
	-.452	-.343	.105	-.598	.134	-.512	.133	-.485	.129	-.438	.129	-.379
	-.311	-.380	.178	-.499	.209	-.447	.214	-.383	.201	-.372	.209	-.303
	-.023	-.332	.286	-.424	.294	-.374	.295	-.349	.294	-.321	.293	-.272
	.133	-.244	.396	-.351	.404	-.321	.407	-.299	.397	-.274	.494	-.219
	.272	-.210	.514	-.285	.497	-.297	.502	-.296	.495	-.283	.590	-.228
	.416	-.170	.618	-.245	.599	-.271	.601	-.290	.594	-.285	.693	-.240
	.565	-.132	.733	-.200	.700	-.284	.698	-.305	.693	-.307	.777	-.254
	.713	-.113	.835	-.167	.864	-.226	.863	-.283	.784	-.304	.861	-.218
	.854	-.133	.919	-.110	.926	-.141	.923	-.182	.856	-.280	.918	-.173
	.980	-.123	.987	-.023	.975	-.051	.977	-.046	.926	-.180	.972	-.038
	1.074	-.109							.977	-.038		
	1.122	-.061										
LOWER SURFACE												
	-.660	.038	-.022	.301	.024	.240	.025	.188	.019	.157	.020	.132
	-.616	.026	.038	.083	.075	-.006	.074	-.001	.066	-.038	.076	-.107
	-.572	.017	.101	-.013	.297	-.078	.130	-.041	.136	-.095	.136	-.156
	-.462	-.034	.185	-.062	.400	-.091	.298	-.070	.214	-.082	.221	-.133
	-.329	-.062	.398	-.100	.604	-.077	.397	-.079	.292	-.081	.295	-.124
	-.172	-.085	.737	.113	.785	.179	.501	-.091	.403	-.068	.396	-.095
	-.030	-.131			.967	.190	.603	.000	.489	-.092	.497	-.083
	.128	-.163			1.000	-.006	.703	.070	.594	-.072	.597	-.029
	.418	-.081					.784	.157	.700	.066	.702	.089
	.564	-.034					.868	.231	.786	.180	.786	.176
	.710	.065					.923	.284	.858	.259	.864	.231
	.976	.223					.972	.180	.919	.270	.912	.239
	1.110	.144							.967	.155	.985	.064
CN=	.3706		.3727		.3938		.3953		.3473		.2699	
CM=	.0144		-.0426		-.0599		-.0783		-.0818		-.0697	

$\alpha = 3.90^{\circ}$; $C_L = 0.408$

	STA .133		STA .307		STA .480		STA .653		STA .804		STA .933	
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.123	-.021	-.718	.023	-1.172	.025	-1.049	.022	-.870	.018	-.694
	-.567	-.267	.035	-.788	.068	-1.126	.079	-.934	.075	-.759	.077	-.659
	-.452	-.359	.105	-.619	.134	-.547	.133	-.531	.129	-.494	.129	-.443
	-.311	-.404	.178	-.552	.209	-.451	.214	-.431	.201	-.415	.209	-.330
	-.023	-.332	.286	-.444	.294	-.395	.295	-.362	.294	-.342	.293	-.288
	.133	-.274	.396	-.369	.404	-.331	.407	-.321	.397	-.300	.494	-.222
	.272	-.212	.514	-.288	.497	-.305	.502	-.312	.495	-.292	.590	-.228
	.416	-.178	.618	-.261	.599	-.290	.601	-.299	.594	-.293	.693	-.248
	.565	-.145	.733	-.214	.700	-.285	.698	-.319	.693	-.306	.777	-.263
	.713	-.118	.835	-.170	.864	-.228	.863	-.271	.784	-.309	.861	-.231
	.854	-.138	.919	-.100	.926	-.138	.923	-.176	.856	-.289	.918	-.178
	.980	-.131	.987	-.012	.975	-.050	.977	-.043	.926	-.176	.972	-.035
	1.074	-.111							.977	-.032		
	1.122	-.060										
LOWER SURFACE												
	-.660	.032	-.022	.326	.024	.300	.025	.227	.019	.237	.020	.226
	-.616	.039	.038	.132	.075	.041	.074	.058	.066	.002	.076	-.031
	-.572	.018	.101	.015	.297	-.059	.130	-.007	.136	-.033	.136	-.110
	-.462	-.018	.185	-.031	.400	-.064	.298	-.039	.214	-.061	.221	-.099
	-.329	-.048	.398	-.072	.604	-.068	.397	-.074	.292	-.051	.295	-.107
	-.172	-.082	.737	.117	.785	.181	.501	-.068	.403	-.054	.396	-.082
	-.030	-.102			.967	.191	.603	.016	.489	-.072	.497	-.075
	.128	-.143			1.000	-.006	.703	.076	.594	-.067	.597	-.026
	.418	-.059					.784	.162	.700	.073	.702	.085
	.564	-.017					.868	.240	.786	.180	.786	.172
	.710	.076					.923	.291	.858	.251	.864	.226
	.976	.234					.972	.181	.919	.269	.912	.232
	1.110	.148							.967	.154	.985	.052
CN=	.4161		.4249		.4381		.4451		.3970		.3146	
CM=	.0191		-.0392		-.0576		-.0762		-.0779		-.0645	

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(c) M = 0.80. Continued.

$\alpha = 4.83^o$; $C_L = 0.502$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.168	-.021	-1.009	.023	-1.415	.025	-1.342	.022	-1.189	.018	-1.067
-.567	-.299	.035	-.944	.068	-1.371	.079	-1.277	.075	-1.126	.077	-.996
-.452	-.389	.105	-.730	.134	-.954	.133	-.680	.129	-.547	.129	-.498
-.311	-.440	.178	-.683	.209	-.475	.214	-.476	.201	-.476	.209	-.384
-.023	-.377	.286	-.494	.294	-.408	.295	-.409	.294	-.389	.293	-.320
.133	-.287	.396	-.412	.404	-.353	.407	-.360	.397	-.344	.494	-.245
.272	-.255	.514	-.315	.497	-.321	.502	-.336	.495	-.317	.590	-.246
.416	-.206	.618	-.277	.599	-.293	.601	-.315	.594	-.315	.693	-.259
.565	-.166	.733	-.218	.700	-.289	.698	-.319	.693	-.320	.777	-.270
.713	-.144	.835	-.162	.864	-.219	.863	-.260	.784	-.311	.861	-.231
.854	-.159	.919	-.084	.926	-.140	.923	-.171	.856	-.273	.918	-.177
.980	-.142	.987	-.017	.975	-.053	.977	-.050	.926	-.160	.972	-.040
1.074	-.127							.977	-.038		
1.122	-.067										
				LOWER SURFACE							
-.660	.039	-.022	.367	.024	.356	.025	.325	.019	.336	.020	.329
-.616	.047	.038	.171	.075	.116	.074	.114	.066	.095	.076	.024
-.572	.044	.101	.070	.297	-.014	.130	.052	.136	.018	.136	-.030
-.462	-.003	.185	.020	.400	-.037	.298	-.022	.214	-.009	.221	-.065
-.329	-.028	.398	-.042	.604	-.051	.397	-.042	.292	-.011	.295	-.071
-.172	-.055	.737	.132	.785	.189	.501	-.053	.403	-.026	.396	-.053
-.030	-.091			.967	.198	.603	.031	.489	-.049	.497	-.061
.128	-.124			1.000	-.002	.703	.084	.594	-.046	.597	-.020
.418	-.034					.784	.160	.700	.076	.702	.087
.564	.006					.868	.248	.786	.179	.786	.164
.710	.095					.923	.296	.858	.261	.864	.215
.976	.246					.972	.187	.919	.275	.912	.225
1.110	.152							.967	.148	.985	.043

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.5009	.5154	.5283	.5288	.4833	.3930
CM=	.0283	-.0351	-.0510	-.0696	-.0719	-.0564

$\alpha = 5.82^o$; $C_L = 0.599$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.224	-.021	-1.463	.023	-1.510	.025	-1.499	.022	-1.280	.018	-1.194
-.567	-.357	.035	-1.191	.068	-1.477	.079	-1.527	.075	-1.239	.077	-1.079
-.452	-.444	.105	-.840	.134	-1.284	.133	-1.168	.129	-.861	.129	-.702
-.311	-.478	.178	-.745	.209	-.874	.214	-.603	.201	-.608	.209	-.454
-.023	-.409	.286	-.562	.294	-.488	.295	-.431	.294	-.479	.293	-.371
.133	-.321	.396	-.438	.404	-.374	.407	-.377	.397	-.394	.494	-.264
.272	-.282	.514	-.326	.497	-.345	.502	-.338	.495	-.352	.590	-.252
.416	-.241	.618	-.289	.599	-.307	.601	-.311	.594	-.321	.693	-.263
.565	-.188	.733	-.212	.700	-.284	.698	-.289	.693	-.306	.777	-.262
.713	-.169	.835	-.154	.864	-.215	.863	-.215	.784	-.281	.861	-.228
.854	-.180	.919	-.078	.926	-.136	.923	-.136	.856	-.242	.918	-.179
.980	-.164	.987	-.030	.975	-.056	.977	-.065	.926	-.159	.972	-.057
1.074	-.137							.977	-.055		
1.122	-.073										
				LOWER SURFACE							
-.660	.034	-.022	.404	.024	.419	.025	.388	.019	.412	.020	.389
-.616	.065	.038	.232	.075	.178	.074	.199	.066	.175	.076	.112
-.572	.068	.101	.126	.297	.030	.130	.115	.136	.089	.136	.016
-.462	.040	.185	.053	.400	-.011	.298	.027	.214	.039	.221	-.016
-.329	-.003	.398	.003	.604	-.030	.397	-.001	.292	.013	.295	-.041
-.172	-.024	.737	.144	.785	.198	.501	-.020	.403	-.000	.396	-.039
-.030	-.053			.967	.199	.603	.051	.489	-.031	.497	-.046
.128	-.086			1.000	.004	.703	.093	.594	-.028	.597	-.014
.418	-.004					.784	.165	.700	.092	.702	.086
.564	.028					.868	.254	.786	.186	.786	.163
.710	.120					.923	.300	.858	.264	.864	.214
.976	.262					.972	.180	.919	.273	.912	.218
1.110	.162							.967	.142	.985	.032

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.6086	.6117	.6348	.6191	.5640	.4543
CM=	.0453	-.0275	-.0474	-.0567	-.0663	-.0520

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; δ_h = -2.5°

(c) M = 0.80. Continued.

$\alpha = 6.96^{\circ}$; $C_L = 0.701$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.254	-.021	-1.625	.023	-1.442	.025	-1.575	.022	-.993	.018	-1.072
-.567	-.411	.035	-1.639	.068	-1.418	.079	-1.498	.075	-.994	.077	-.924
-.452	-.487	.105	-1.071	.134	-1.224	.133	-1.239	.129	-.926	.129	-.800
-.311	-.510	.178	-.915	.209	-1.071	.214	-.706	.201	-.825	.209	-.631
-.023	-.438	.286	-.668	.294	-.916	.295	-.496	.294	-.740	.293	-.476
.133	-.343	.396	-.464	.404	-.725	.407	-.409	.397	-.628	.494	-.283
.272	-.327	.514	-.349	.497	-.507	.502	-.350	.495	-.485	.590	-.240
.416	-.290	.618	-.300	.599	-.370	.601	-.294	.594	-.387	.693	-.214
.565	-.240	.733	-.229	.700	-.309	.698	-.255	.693	-.327	.777	-.195
.713	-.214	.835	-.161	.864	-.216	.863	-.212	.784	-.257	.861	-.156
.854	-.204	.919	-.089	.926	-.161	.923	-.171	.856	-.217	.918	-.135
.980	-.190	.987	-.035	.975	-.083	.977	-.149	.926	-.166	.972	-.091
1.074	-.149							.977	-.105		
1.122	-.092										
				LOWER SURFACE							
-.660	.034	-.022	.426	.024	.466	.025	.451	.019	.447	.020	.439
-.616	.079	.038	.307	.075	.263	.074	.239	.066	.220	.076	.173
-.572	.082	.101	.198	.297	.061	.130	.146	.136	.124	.136	.063
-.462	.057	.185	.115	.400	.031	.298	.058	.214	.086	.221	-.004
-.329	.032	.398	.037	.604	-.010	.397	.019	.292	.051	.295	-.014
-.172	.001	.737	.152	.785	.202	.501	-.006	.403	.019	.396	-.035
-.030	-.032			.967	.195	.603	.059	.489	-.010	.497	-.045
.128	-.057			1.000	-.013	.703	.079	.594	-.018	.597	-.023
.418	.023					.784	.149	.700	.093	.702	.075
.564	.062					.868	.244	.786	.187	.786	.149
.710	.141					.923	.298	.858	.260	.864	.198
.976	.279					.972	.155	.919	.269	.912	.195
1.110	.167							.967	.130	.985	-.014
CN=	.7150		.7378		.7751		.6577		.6468		.4670
CM=	.0506		-.0202		-.0658		-.0549		-.0806		-.0400

$\alpha = 7.99^{\circ}$; $C_L = 0.779$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.296	-.021	-1.695	.023	-1.248	.025	-1.489	.022	-.951	.018	-.646
-.567	-.446	.035	-1.611	.068	-1.190	.079	-1.455	.075	-.953	.077	-.623
-.452	-.533	.105	-1.305	.134	-1.061	.133	-1.168	.129	-.916	.129	-.573
-.311	-.528	.178	-1.171	.209	-1.088	.214	-.657	.201	-.886	.209	-.541
-.023	-.465	.286	-.942	.294	-.902	.295	-.471	.294	-.829	.293	-.491
.133	-.376	.396	-.529	.404	-.861	.407	-.406	.397	-.758	.494	-.400
.272	-.338	.514	-.362	.497	-.768	.502	-.357	.495	-.651	.590	-.349
.416	-.310	.618	-.294	.599	-.653	.601	-.300	.594	-.537	.693	-.295
.565	-.267	.733	-.212	.700	-.488	.698	-.274	.693	-.450	.777	-.257
.713	-.225	.835	-.161	.864	-.356	.863	-.234	.784	-.352	.861	-.225
.854	-.238	.919	-.092	.926	-.285	.923	-.237	.856	-.275	.918	-.204
.980	-.193	.987	-.050	.975	-.205	.977	-.224	.926	-.213	.972	-.191
1.074	-.152							.977	-.179		
1.122	-.084										
				LOWER SURFACE							
-.660	.029	-.022	.428	.024	.512	.025	.469	.019	.468	.020	.452
-.616	.103	.038	.341	.075	.299	.074	.267	.066	.245	.076	.190
-.572	.109	.101	.245	.297	.089	.130	.178	.136	.139	.136	.079
-.462	.078	.185	.165	.400	.054	.298	.068	.214	.086	.221	.014
-.329	.054	.398	.063	.604	-.004	.397	.027	.292	.061	.295	-.014
-.172	.031	.737	.163	.785	.198	.501	-.013	.403	.030	.396	-.035
-.030	.001			.967	.139	.603	.037	.489	-.013	.497	-.064
.128	-.015			1.000	-.113	.703	.056	.594	-.023	.597	-.047
.418	.058					.784	.124	.700	.086	.702	.041
.564	.081					.868	.226	.786	.181	.786	.115
.710	.160					.923	.277	.858	.251	.864	.160
.976	.288					.972	.115	.919	.252	.912	.159
1.110	.175							.967	.094	.985	-.111
CN=	.8010		.8440		.8712		.6496		.7301		.4493
CM=	.0634		-.0195		-.1137		-.0583		-.1054		-.0620

[redacted]

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Continued.

$\alpha = 8.99^{\circ}$; $C_L = 0.830$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.364	-.021	-1.622	.023	-.904	.025	-1.422	.022	-1.000	.018	-.608
-.567	-.494	.035	-1.579	.068	-.890	.079	-1.240	.075	-.985	.077	-.592
-.452	-.561	.105	-1.489	.134	-.867	.133	-1.031	.129	-.969	.129	-.554
-.311	-.575	.178	-1.425	.209	-.796	.214	-.552	.201	-.941	.209	-.514
-.023	-.492	.286	-1.197	.294	-.800	.295	-.434	.294	-.900	.293	-.506
.133	-.398	.396	-.759	.404	-.735	.407	-.360	.397	-.811	.494	-.419
.272	-.367	.514	-.360	.497	-.706	.502	-.346	.495	-.717	.590	-.380
.416	-.328	.618	-.252	.599	-.648	.601	-.322	.594	-.605	.693	-.331
.565	-.274	.733	-.209	.700	-.603	.698	-.323	.693	-.537	.777	-.312
.713	-.255	.835	-.172	.864	-.503	.863	-.297	.784	-.415	.861	-.275
.854	-.250	.919	-.128	.926	-.506	.923	-.304	.856	-.362	.918	-.262
.980	-.215	.987	-.047	.975	-.443	.977	-.301	.926	-.294	.972	-.238
1.074	-.160							.977	-.188		
1.122	-.102										
					LOWER SURFACE						
-.660	.014	-.022	.434	.024	.538	.025	.491	.019	.480	.020	.462
-.616	.105	.038	.368	.075	.343	.074	.274	.066	.274	.076	.206
-.572	.132	.101	.277	.297	.117	.130	.188	.136	.161	.136	.103
-.462	.111	.185	.203	.400	.068	.298	.068	.214	.108	.221	.031
-.329	.078	.398	.093	.604	-.020	.397	.020	.292	.067	.295	-.002
-.172	.060	.737	.172	.785	.142	.501	-.036	.403	.032	.396	-.037
-.030	.037			.967	.015	.603	.020	.489	-.014	.497	-.066
.128	.015			1.000	-.484	.703	.033	.594	-.026	.597	-.063
.418	.072					.784	.114	.700	.086	.702	.025
.564	.110					.868	.213	.786	.185	.786	.086
.710	.180					.923	.262	.858	.252	.864	.155
.976	.297					.972	.091	.919	.257	.912	.140
1.110	.179							.967	.100	.985	-.149
CN=	.8894		.9516		.8145		.6227		.7993		.4639
CM=	.0783		-.0246		-.1349		-.0696		-.1226		-.0695

$\alpha = 9.97^{\circ}$; $C_L = 0.882$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.415	-.021	-1.556	.023	-.725	.025	-1.447	.022	-.985	.018	-.565
-.567	-.551	.035	-1.512	.068	-.738	.079	-1.287	.075	-.975	.077	-.561
-.452	-.604	.105	-1.485	.134	-.710	.133	-1.153	.129	-.964	.129	-.542
-.311	-.589	.178	-1.479	.209	-.714	.214	-.475	.201	-.964	.209	-.519
-.023	-.521	.286	-1.354	.294	-.692	.295	-.402	.294	-.940	.293	-.493
.133	-.427	.396	-1.121	.404	-.675	.407	-.347	.397	-.861	.494	-.413
.272	-.388	.514	-.306	.497	-.615	.502	-.342	.495	-.811	.590	-.387
.416	-.341	.618	-.380	.599	-.613	.601	-.340	.594	-.707	.693	-.340
.565	-.285	.733	-.271	.700	-.571	.698	-.334	.693	-.592	.777	-.309
.713	-.236	.835	-.272	.864	-.539	.863	-.317	.784	-.496	.861	-.282
.854	-.244	.919	-.196	.926	-.501	.923	-.298	.856	-.419	.918	-.274
.980	-.227	.987	-.098	.975	-.476	.977	-.296	.926	-.335	.972	-.264
1.074	-.169							.977	-.225		
1.122	-.099										
					LOWER SURFACE						
-.660	.000	-.022	.424	.024	.543	.025	.494	.019	.491	.020	.487
-.616	.116	.038	.404	.075	.377	.074	.301	.066	.302	.076	.236
-.572	.137	.101	.312	.297	.122	.130	.207	.136	.193	.136	.123
-.462	.139	.185	.219	.400	.071	.298	.067	.214	.124	.221	.036
-.329	.109	.398	.100	.604	-.040	.397	.017	.292	.087	.295	.001
-.172	.079	.737	.162	.785	.111	.501	-.033	.403	.029	.396	-.039
-.030	.054			.967	-.038	.603	.010	.489	-.008	.497	-.078
.128	.031			1.000	-.540	.703	.024	.594	-.019	.597	-.070
.418	.092					.784	.099	.700	.082	.702	.005
.564	.114					.868	.216	.786	.182	.786	.074
.710	.188					.923	.258	.858	.252	.864	.138
.976	.303					.972	.071	.919	.251	.912	.132
1.110	.169							.967	.093	.985	-.198
CN=	.9463		1.0441		.7426		.6281		.8554		.4568
CM=	.0993		-.0477		-.1270		-.0672		-.1397		-.0668

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Concluded.

$\alpha = 11.01^{\circ}$; $C_L = 0.949$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.459	-.021	-1.507	.023	-.828	.025	-1.446	.022	-.985	.018	-.536
-.567	-.591	.035	-1.459	.068	-.811	.079	-1.393	.075	-.975	.077	-.526
-.452	-.642	.105	-1.479	.134	-.804	.133	-1.252	.129	-.950	.129	-.526
-.311	-.639	.178	-1.386	.209	-.765	.214	-.332	.201	-.991	.209	-.493
-.023	-.523	.286	-1.268	.294	-.747	.295	-.354	.294	-.965	.293	-.448
.133	-.455	.396	-1.109	.404	-.700	.407	-.329	.397	-.941	.494	-.421
.272	-.406	.514	-.976	.497	-.659	.502	-.338	.495	-.881	.590	-.388
.416	-.376	.618	-.788	.599	-.623	.601	-.323	.594	-.808	.693	-.357
.565	-.297	.733	-.576	.700	-.574	.698	-.337	.693	-.694	.777	-.342
.713	-.267	.835	-.371	.864	-.527	.863	-.315	.784	-.611	.861	-.320
.854	-.291	.919	-.311	.926	-.498	.923	-.315	.856	-.447	.918	-.309
.980	-.270	.987	-.227	.975	-.450	.977	-.318	.926	-.332	.972	-.309
1.074	-.207							.977	-.195		
1.122	-.124										
					LOWER SURFACE						
-.660	-.014	-.022	.415	.024	.564	.025	.516	.019	.517	.020	.487
-.616	.124	.038	.439	.075	.399	.074	.342	.066	.327	.076	.255
-.572	.165	.101	.355	.297	.146	.130	.239	.136	.209	.136	.145
-.462	.179	.185	.273	.400	.085	.298	.091	.214	.147	.221	.056
-.329	.140	.398	.142	.604	-.039	.397	.043	.292	.104	.295	.021
-.172	.107	.737	.173	.785	.108	.501	-.015	.403	.054	.396	-.028
-.030	.087			.967	-.029	.603	.019	.489	-.005	.497	-.065
.128	.061			1.000	-.483	.703	.028	.594	-.028	.597	-.067
.418	.126					.784	.098	.700	.080	.702	.012
.564	.136					.868	.215	.786	.169	.786	.072
.710	.216					.923	.256	.858	.247	.864	.133
.976	.319					.972	.060	.919	.248	.912	.134
1.110	.178							.967	.078	.985	-.191
CN=	1.0492		1.2294		.7838		.6375		.9098		.4659
CM=	.1047		-.1207		-.1250		-.0661		-.1533		-.0751

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90.

$\alpha = -5.00^{\circ}$; $C_L = -0.514$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.038	-.021	.349	.023	.369	.025	.365	.022	.456	.018	.469
-.567	-.005	.035	.197	.068	.235	.079	.232	.075	.236	.077	.216
-.452	-.106	.105	.103	.209	.084	.133	.169	.129	.183	.129	.157
-.311	-.178	.178	.065	.294	.038	.214	.099	.201	.126	.209	.087
-.023	-.098	.286	.014	.404	.015	.295	.066	.294	.078	.293	.015
.133	-.026	.396	-.002	.497	-.011	.407	.020	.397	.032	.494	-.094
.272	.001	.514	-.007	.599	-.047	.502	-.032	.495	-.030	.590	-.163
.416	.035	.618	-.029	.700	-.103	.601	-.088	.594	-.086	.693	-.260
.565	.063	.733	-.050	.864	-.203	.698	-.157	.693	-.170	.777	-.363
.713	.065	.835	-.074	.926	-.189	.863	-.306	.784	-.225	.861	-.483
.854	.018	.919	-.067			.923	-.297	.856	-.269	.918	-.630
.980	-.014	.987	.025			.977	-.132	.926	-.217	.972	-.383
1.074	-.041							.977	-.155		

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.068	-.022	-.762	.024	-1.165	.074	-1.291	.019	-.747	.020	-.371
-.616	-.157	.038	-.986	.075	-1.194	.130	-1.258	.066	-.734	.076	-.365
-.462	-.228	.101	-.665	.297	-1.042	.298	-.732	.136	-.699	.136	-.343
-.329	-.224	.185	-.676	.400	-.691	.397	-.320	.214	-.682	.221	-.335
-.172	-.249	.398	-.691	.604	-.146	.501	-.254	.292	-.678	.295	-.327
-.030	-.304	.737	-.108	.785	.102	.603	-.143	.403	-.685	.396	-.289
.128	-.351			.967	.108	.784	-.054	.489	-.659	.497	-.255
.418	-.369			1.000	.001	.868	-.010	.594	-.595	.597	-.260
.564	-.396					.923	-.010	.700	-.552	.702	-.247
.710	-.283					.972	-.002	.786	-.485	.786	-.240
.976	.097							.858	-.415	.864	-.238
1.072	.151							.919	-.357	.912	-.225
1.110	.129							.967	-.293	.985	-.227

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	-.3652	-.5455	-.4823	-.4270	-.5738	-.1627
CM=	-.0233	-.0061	-.0686	-.0698	.0578	-.0463

$\alpha = -3.98^{\circ}$; $C_L = -0.428$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.030	-.021	.311	.023	.308	.025	.313	.022	.415	.018	.443
-.567	-.030	.035	.111	.068	.174	.079	.177	.075	.193	.077	.175
-.452	-.146	.105	.051	.209	.038	.133	.112	.129	.162	.129	.139
-.311	-.203	.178	.009	.294	.007	.214	.068	.201	.098	.209	.065
-.023	-.132	.286	-.025	.404	-.024	.295	.034	.294	.057	.293	.004
.133	-.053	.396	-.040	.497	-.049	.407	-.002	.397	.015	.494	-.104
.272	-.020	.514	-.041	.599	-.079	.502	-.042	.495	-.041	.590	-.159
.416	.014	.618	-.059	.700	-.123	.601	-.092	.594	-.093	.693	-.241
.565	.045	.733	-.078	.864	-.188	.698	-.156	.693	-.169	.777	-.354
.713	.035	.835	-.095	.926	-.158	.863	-.256	.784	-.224	.861	-.458
.854	-.004	.919	-.084			.923	-.231	.856	-.231	.918	-.564
.980	-.025	.987	.013			.977	-.106	.926	-.149	.972	-.380
1.074	-.055							.977	-.028		

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.035	-.022	-.626	.024	-1.098	.074	-1.299	.019	-.817	.020	-.450
-.616	-.132	.038	-.763	.075	-1.104	.130	-1.217	.066	-.769	.076	-.407
-.462	-.203	.101	-.641	.297	-1.039	.298	-.712	.136	-.761	.136	-.361
-.329	-.213	.185	-.627	.400	-.496	.397	-.271	.214	-.714	.221	-.328
-.172	-.238	.398	-.629	.604	-.151	.501	-.197	.292	-.707	.295	-.313
-.030	-.290	.737	-.044	.785	.091	.603	-.113	.403	-.680	.396	-.285
.128	-.382			.967	.149	.784	.031	.489	-.638	.497	-.243
.418	-.351			1.000	.031	.868	.081	.594	-.577	.597	-.228
.564	-.359					.923	.059	.700	-.505	.702	-.222
.710	-.204					.972	.036	.786	-.401	.786	-.217
.976	.124							.858	-.265	.864	-.199
1.072	.160							.919	-.197	.912	-.202
1.110	.133							.967	-.083	.985	-.202

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	-.2742	-.4300	-.4102	-.3745	-.5418	-.1530
CM=	-.0250	-.0211	-.0710	-.0792	.0382	-.0493

[redacted]

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(d) M = 0.90. Continued.

$\alpha = -3.13^o; C_L = -0.345$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	.025	-.021	.282	.023	.237	.025	.257	.022	.377	.018	.419
-.567	-.059	.035	.065	.068	.096	.079	.135	.075	.143	.077	.136
-.452	-.157	.105	.003	.209	-.011	.133	.076	.129	.123	.129	.103
-.311	-.225	.178	-.035	.294	-.030	.214	.031	.201	.060	.209	.049
-.023	-.159	.286	-.070	.404	-.050	.295	.014	.294	.027	.293	-.018
.133	-.075	.396	-.084	.497	-.075	.407	-.033	.397	-.011	.494	-.103
.272	-.043	.514	-.064	.599	-.096	.502	-.070	.495	-.059	.590	-.157
.416	-.012	.618	-.082	.700	-.140	.601	-.104	.594	-.109	.693	-.221
.565	.020	.733	-.094	.864	-.176	.698	-.169	.693	-.175	.777	-.301
.713	.014	.835	-.110	.926	-.124	.863	-.245	.784	-.223	.861	-.382
.854	-.014	.919	-.089			.923	-.192	.856	-.234	.918	-.426
.980	-.040	.987	.004			.977	-.051	.926	-.159	.972	-.366
1.074	-.060							.977	-.007		
				LOWER SURFACE							
-.660	-.020	-.022	-.423	.024	-.958	.074	-1.221	.019	-.872	.020	-.505
-.616	-.104	.038	-.607	.075	-.993	.130	-1.138	.066	-.862	.076	-.472
-.462	-.187	.101	-.585	.297	-.969	.298	-.578	.136	-.813	.136	-.415
-.329	-.202	.185	-.562	.400	-.154	.397	-.223	.214	-.780	.221	-.367
-.172	-.220	.398	-.595	.604	-.178	.501	-.154	.292	-.711	.295	-.330
-.030	-.278	.737	.002	.785	.121	.603	-.026	.403	-.637	.396	-.302
.128	-.354			.967	.206	.784	.061	.489	-.545	.497	-.258
.418	-.326			1.000	.037	.868	.113	.594	-.467	.597	-.238
.564	-.323					.923	.115	.700	-.343	.702	-.220
.710	-.153					.972	.112	.786	-.205	.786	-.221
.976	.136							.858	-.058	.864	-.182
1.072	.167							.919	.015	.912	-.189
1.110	.137							.967	.092	.985	-.175
CN=	-.1968		-.3331		-.2943		-.2900		-.4487		-.1795
CM=	-.0236		-.0292		-.0783		-.0888		-.0094		-.0391

$\alpha = -2.12^o; C_L = -0.234$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	.022	-.021	.211	.023	.148	.025	.198	.022	.314	.018	.378
-.567	-.082	.035	-.021	.068	.039	.079	.046	.075	.065	.077	.089
-.452	-.193	.105	-.061	.209	-.070	.133	.019	.129	.060	.129	.060
-.311	-.261	.178	-.098	.294	-.087	.214	-.018	.201	.013	.209	.006
-.023	-.192	.286	-.133	.404	-.100	.295	-.041	.294	-.011	.293	-.054
.133	-.113	.396	-.137	.497	-.117	.407	-.072	.397	-.042	.494	-.116
.272	-.067	.514	-.109	.599	-.141	.502	-.104	.495	-.088	.590	-.151
.416	-.037	.618	-.113	.700	-.172	.601	-.134	.594	-.134	.693	-.190
.565	-.008	.733	-.117	.864	-.195	.698	-.192	.693	-.197	.777	-.229
.713	.000	.835	-.134	.926	-.131	.863	-.244	.784	-.233	.861	-.202
.854	-.040	.919	-.105			.923	-.167	.856	-.254	.918	-.224
.980	-.060	.987	-.002			.977	-.019	.926	-.183	.972	-.155
1.074	-.070							.977	-.017		
				LOWER SURFACE							
-.660	-.020	-.022	-.224	.024	-.848	.074	-1.098	.019	-.912	.020	-.828
-.616	-.060	.038	-.515	.075	-.788	.130	-1.032	.066	-.896	.076	-.805
-.462	-.165	.101	-.502	.297	-.376	.298	-.328	.136	-.696	.136	-.632
-.329	-.187	.185	-.491	.400	-.287	.397	-.260	.214	-.590	.221	-.540
-.172	-.208	.398	-.415	.604	-.197	.501	-.181	.292	-.512	.295	-.479
-.030	-.264	.737	.036	.785	.127	.603	-.030	.403	-.454	.396	-.371
.128	-.346			.967	.214	.784	.118	.489	-.284	.497	-.220
.418	-.283			1.000	.042	.868	.163	.594	-.179	.597	-.164
.564	-.271					.923	.186	.700	-.070	.702	-.068
.710	-.065					.972	.148	.786	-.002	.786	-.024
.976	.150							.858	.082	.864	-.026
1.072	.168							.919	.118	.912	.046
1.110	.139							.967	.091	.985	.031
CN=	-.1015		-.1864		-.1422		-.1917		-.2464		-.2141
CM=	-.0251		-.0422		-.0792		-.0951		-.0620		-.0595

[redacted]

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(d) M = 0.90. Continued.

$\alpha = -1.16^o$; $C_L = -0.126$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.007	-.021	.104	.023	.035	.025	.084	.022	.231	.018	.321
-.567	-.112	.035	-.111	.068	-.072	.079	-.054	.075	-.006	.077	.019
-.452	-.225	.105	-.162	.209	-.134	.133	-.064	.129	-.009	.129	-.003
-.311	-.307	.178	-.158	.294	-.145	.214	-.069	.201	-.044	.209	-.042
-.023	-.228	.286	-.193	.404	-.147	.295	-.090	.294	-.061	.293	-.088
.133	-.134	.396	-.175	.497	-.152	.407	-.109	.397	-.075	.494	-.140
.272	-.104	.514	-.143	.599	-.168	.502	-.137	.495	-.126	.590	-.164
.416	-.070	.618	-.155	.700	-.197	.601	-.171	.594	-.161	.693	-.190
.565	-.021	.733	-.150	.864	-.210	.698	-.227	.693	-.221	.777	-.204
.713	-.021	.835	-.154	.926	-.137	.863	-.255	.784	-.255	.861	-.175
.854	-.058	.919	-.111			.923	-.163	.856	-.264	.918	-.138
.980	-.077	.987	-.005			.977	-.010	.926	-.180	.972	-.001
1.074	-.079							.977	-.010		

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.004	-.022	-.111	.024	-.591	.074	-.910	.019	-.883	.020	-.946
-.616	-.069	.038	-.345	.075	-.644	.130	-.877	.066	-1.010	.076	-.869
-.462	-.155	.101	-.425	.297	-.336	.298	-.289	.136	-.936	.136	-.781
-.329	-.168	.185	-.422	.400	-.268	.397	-.296	.214	-.457	.221	-.545
-.172	-.197	.398	-.334	.604	-.161	.501	-.217	.292	-.305	.295	-.363
-.030	-.243	.737	.059	.785	.124	.603	-.047	.403	-.241	.396	-.176
.128	-.316			.967	.213	.784	.124	.489	-.237	.497	-.096
.418	-.250			1.000	.036	.868	.186	.594	-.154	.597	-.031
.564	-.206					.923	.201	.700	.018	.702	.049
.710	-.053					.972	.152	.786	.107	.786	.106
.976	.167							.858	.185	.864	.137
1.072	.180							.919	.197	.912	.146
1.110	.141							.967	.144	.985	.089

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	-.0073	-.0697	-.0495	-.1148	-.1444	-.1309
CM=	-.0213	-.0479	-.0784	-.0918	-.0904	-.0849

$\alpha = 0.02^o$; $C_L = 0.008$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.015	-.021	.006	.023	-.160	.025	-.047	.022	.085	.018	.195
-.567	-.136	.035	-.212	.068	-.167	.079	-.149	.075	-.140	.077	-.086
-.452	-.253	.105	-.241	.209	-.223	.133	-.157	.129	-.108	.129	-.082
-.311	-.353	.178	-.243	.294	-.208	.214	-.164	.201	-.125	.209	-.112
-.023	-.253	.286	-.246	.404	-.191	.295	-.164	.294	-.123	.293	-.147
.133	-.167	.396	-.238	.497	-.194	.407	-.169	.397	-.134	.494	-.165
.272	-.127	.514	-.178	.599	-.198	.502	-.186	.495	-.167	.590	-.181
.416	-.104	.618	-.186	.700	-.230	.601	-.207	.594	-.202	.693	-.198
.565	-.052	.733	-.173	.864	-.220	.698	-.256	.693	-.253	.777	-.210
.713	-.052	.835	-.167	.926	-.138	.863	-.268	.784	-.283	.861	-.173
.854	-.082	.919	-.116			.923	-.161	.856	-.280	.918	-.124
.980	-.089	.987	-.004			.977	-.008	.926	-.172	.972	.011
1.074	-.087							.977	-.013		

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.012	-.022	.004	.024	-.309	.074	-.570	.019	-.646	.020	-.627
-.616	-.042	.038	-.257	.075	-.448	.130	-.293	.066	-.722	.076	-.772
-.462	-.112	.101	-.320	.297	-.292	.298	-.298	.136	-.526	.136	-.714
-.329	-.150	.185	-.308	.400	-.235	.397	-.250	.214	-.325	.221	-.378
-.172	-.181	.398	-.266	.604	-.123	.501	-.192	.292	-.281	.295	-.243
-.030	-.224	.737	.076	.785	.147	.603	-.042	.403	-.213	.396	-.165
.128	-.286			.967	.211	.784	.120	.489	-.217	.497	-.126
.418	-.209			1.000	.018	.868	.183	.594	-.160	.597	-.039
.564	-.159					.923	.190	.700	.023	.702	.098
.710	-.021					.972	.146	.786	.133	.786	.162
.976	.192							.858	.188	.864	.204
1.072	.186							.919	.223	.912	.207
1.110	.147							.967	.159	.985	.094

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.0954	.0473	.0648	.0427	-.0120	-.0303
CM=	-.0153	-.0510	-.0783	-.0811	-.0875	-.0870

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90. Continued.

$\alpha = 1.00^{\circ}$; $C_L = 0.115$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.026	-.021	-.145	.023	-.364	.025	-.226	.022	-.059	.018	.075
-.567	-.174	.035	-.335	.068	-.404	.079	-.298	.075	-.253	.077	-.163
-.452	-.287	.105	-.345	.209	-.278	.133	-.255	.129	-.207	.129	-.160
-.311	-.380	.178	-.325	.294	-.274	.214	-.227	.201	-.193	.209	-.176
-.023	-.277	.286	-.318	.404	-.246	.295	-.207	.294	-.186	.293	-.183
.133	-.190	.396	-.295	.497	-.236	.407	-.209	.397	-.183	.494	-.188
.272	-.160	.514	-.213	.599	-.234	.502	-.220	.495	-.208	.590	-.197
.416	-.133	.618	-.213	.700	-.253	.601	-.240	.594	-.235	.693	-.214
.565	-.085	.733	-.190	.864	-.230	.698	-.285	.693	-.277	.777	-.226
.713	-.074	.835	-.171	.926	-.138	.863	-.261	.784	-.301	.861	-.186
.854	-.101	.919	-.115			.923	-.152	.856	-.284	.918	-.138
.980	-.110	.987	-.009			.977	-.013	.926	-.168	.972	.002
1.074	-.097							.977	-.019		
					LOWER SURFACE						
-.660	.023	-.022	.117	.024	-.101	.074	-.333	.019	-.376	.020	-.335
-.616	-.019	.038	-.130	.075	-.299	.130	-.292	.066	-.445	.076	-.479
-.462	-.093	.101	-.224	.297	-.243	.298	-.221	.136	-.353	.136	-.496
-.329	-.132	.185	-.244	.400	-.192	.397	-.201	.214	-.276	.221	-.379
-.172	-.153	.398	-.223	.604	-.113	.501	-.156	.292	-.204	.295	-.186
-.030	-.188	.737	.094	.785	.165	.603	-.030	.403	-.181	.396	-.164
.128	-.243			.967	.210	.784	.149	.489	-.187	.497	-.125
.418	-.159			1.000	.003	.868	.194	.594	-.150	.597	-.036
.564	-.119					.923	.211	.700	.045	.702	.102
.710	.003					.972	.169	.786	.161	.786	.183
.976	.204							.858	.219	.864	.229
1.072	.205							.919	.251	.912	.239
1.110	.147							.967	.163	.985	.089
CN=	.1941		.1530		.1757		.1412		.1017		.0629
CM=	-.0095		-.0508		-.0755		-.0831		-.0879		-.0841

$\alpha = 1.99^{\circ}$; $C_L = 0.224$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.048	-.021	-.300	.023	-.563	.025	-.424	.022	-.279	.018	-.136
-.567	-.207	.035	-.461	.069	-.790	.079	-.441	.075	-.423	.077	-.275
-.452	-.307	.105	-.445	.209	-.406	.133	-.397	.129	-.334	.129	-.276
-.311	-.404	.178	-.404	.294	-.307	.214	-.317	.201	-.279	.209	-.247
-.023	-.310	.286	-.352	.404	-.293	.295	-.279	.294	-.251	.293	-.253
.133	-.224	.396	-.355	.497	-.306	.407	-.265	.397	-.229	.494	-.214
.272	-.181	.514	-.240	.599	-.252	.502	-.264	.495	-.248	.590	-.216
.416	-.154	.618	-.241	.700	-.278	.601	-.286	.594	-.266	.693	-.233
.565	-.139	.733	-.204	.864	-.226	.698	-.316	.693	-.298	.777	-.246
.713	-.098	.835	-.175	.926	-.128	.863	-.266	.784	-.317	.861	-.200
.854	-.132	.919	-.106			.923	-.157	.856	-.285	.918	-.145
.980	-.124	.987	-.009			.977	-.028	.926	-.160	.972	-.003
1.074	-.103							.977	-.025		
					LOWER SURFACE						
-.660	.032	-.022	.203	.024	.018	.074	-.195	.019	-.127	.020	-.105
-.616	-.006	.038	-.044	.075	-.164	.130	-.198	.066	-.272	.076	-.320
-.462	-.078	.101	-.127	.297	-.182	.298	-.160	.136	-.226	.136	-.360
-.329	-.109	.185	-.154	.400	-.159	.397	-.160	.214	-.194	.221	-.279
-.172	-.141	.398	-.166	.604	-.102	.501	-.142	.292	-.173	.295	-.222
-.030	-.184	.737	.104	.785	.175	.603	-.022	.403	-.125	.396	-.141
.128	-.223			.967	.208	.784	.151	.489	-.148	.497	-.119
.418	-.128			1.000	.008	.868	.221	.594	-.132	.597	-.039
.564	-.085					.923	.256	.700	.057	.702	.100
.710	.030					.972	.182	.786	.173	.786	.188
.976	.214							.858	.248	.864	.239
1.072	.202							.919	.268	.912	.246
1.110	.155							.967	.163	.985	.078
CN=	.2732		.2554		.2960		.2483		.2135		.1514
CM=	-.0069		-.0491		-.0687		-.0863		-.0870		-.0794

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90. Continued.

$\alpha = 2.46^{\circ}$; $C_L = 0.275$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.063	-.021	-.356	.023	-.691	.025	-.497	.022	-.350	.018	-.229
-.567	-.227	.035	-.548	.068	-.822	.079	-.737	.075	-.532	.077	-.411
-.452	-.329	.105	-.458	.209	-.464	.133	-.453	.129	-.372	.129	-.330
-.311	-.403	.178	-.470	.294	-.318	.214	-.367	.201	-.332	.209	-.293
-.023	-.307	.286	-.403	.404	-.269	.295	-.311	.294	-.267	.293	-.281
.133	-.233	.396	-.376	.497	-.314	.407	-.283	.397	-.266	.494	-.219
.272	-.197	.514	-.325	.599	-.265	.502	-.289	.495	-.280	.590	-.222
.416	-.174	.618	-.247	.700	-.287	.601	-.288	.594	-.282	.693	-.240
.565	-.130	.733	-.205	.864	-.230	.698	-.332	.693	-.321	.777	-.246
.713	-.114	.835	-.173	.926	-.127	.863	-.271	.784	-.329	.861	-.202
.854	-.142	.919	-.109			.923	-.153	.856	-.284	.918	-.147
.980	-.129	.987	-.011			.977	-.034	.926	-.152	.972	-.002
1.074	-.113							.977	-.028		
				LOWER SURFACE							
-.660	.036	-.022	.241	.024	.095	.074	-.123	.019	-.028	.020	-.007
-.616	.012	.038	-.012	.075	-.128	.130	-.162	.066	-.196	.076	-.251
-.462	-.067	.101	-.088	.297	-.162	.298	-.127	.136	-.177	.136	-.274
-.329	-.094	.185	-.139	.400	-.139	.397	-.151	.214	-.165	.221	-.241
-.172	-.140	.398	-.142	.604	-.091	.501	-.116	.292	-.157	.295	-.203
-.030	-.173	.737	.111	.785	.183	.603	-.012	.403	-.106	.396	-.123
.128	-.205			.967	.206	.784	.157	.489	-.144	.497	-.112
.418	-.117			1.000	.000	.868	.227	.594	-.121	.597	-.034
.564	-.074					.923	.275	.700	.059	.702	.098
.710	.043					.972	.186	.786	.179	.786	.187
.976	.215							.858	.249	.864	.239
1.072	.204							.919	.271	.912	.245
1.110	.153							.967	.161	.985	.072
CN=	.3098		.3082		.3319		.3082		.2627		.1994
CM=	-.0043		-.0511		-.0679		-.0858		-.0868		-.0752

$\alpha = 2.98^{\circ}$; $C_L = 0.332$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.084	-.021	-.475	.023	-.779	.025	-.731	.022	-.497	.018	-.374
-.567	-.234	.035	-.681	.068	-.917	.079	-.834	.075	-.784	.077	-.534
-.452	-.334	.105	-.508	.209	-.551	.133	-.539	.129	-.436	.129	-.407
-.311	-.423	.178	-.513	.294	-.423	.214	-.373	.201	-.404	.209	-.331
-.023	-.293	.286	-.428	.404	-.269	.295	-.330	.294	-.333	.293	-.316
.133	-.264	.396	-.391	.497	-.296	.407	-.318	.397	-.289	.494	-.222
.272	-.204	.514	-.324	.599	-.263	.502	-.311	.495	-.296	.590	-.222
.416	-.186	.618	-.257	.700	-.284	.601	-.306	.594	-.301	.693	-.243
.565	-.148	.733	-.204	.864	-.229	.698	-.339	.693	-.327	.777	-.248
.713	-.138	.835	-.161	.926	-.129	.863	-.270	.784	-.327	.861	-.205
.854	-.157	.919	-.092			.923	-.155	.856	-.275	.918	-.148
.980	-.144	.987	-.008			.977	-.037	.926	-.145	.972	-.009
1.074	-.115							.977	-.029		
				LOWER SURFACE							
-.660	.038	-.022	.276	.024	.164	.074	-.058	.019	.092	.020	.107
-.616	.016	.038	.032	.075	-.077	.130	-.093	.066	-.112	.076	-.187
-.462	-.055	.101	-.054	.297	-.124	.298	-.112	.136	-.139	.136	-.229
-.329	-.086	.185	-.105	.400	-.120	.397	-.117	.214	-.135	.221	-.193
-.172	-.119	.398	-.130	.604	-.081	.501	-.100	.292	-.124	.295	-.182
-.030	-.158	.737	.116	.785	.182	.603	-.004	.403	-.090	.396	-.119
.128	-.194			.967	.205	.784	.155	.489	-.120	.497	-.104
.418	-.104			1.000	-.001	.868	.236	.594	-.108	.597	-.033
.564	-.056					.923	.284	.700	.062	.702	.099
.710	.054					.972	.190	.786	.178	.786	.184
.976	.230							.858	.254	.864	.233
1.072	.202							.919	.273	.912	.242
1.110	.160							.967	.158	.985	.066
CN=	.3519		.3517		.3835		.3672		.3291		.2458
CM=	-.0052		-.0458		-.0642		-.0838		-.0824		-.0699

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90. Continued.

$\alpha = 3.43^{\circ}$; $C_L = 0.383$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.080	-.021	-.692	.023	-.859	.025	-.818	.022	-.667	.018	-.535
-.567	-.258	.035	-.663	.068	-1.016	.079	-.915	.075	-.944	.077	-.800
-.452	-.349	.105	-.571	.209	-.657	.133	-.846	.129	-.590	.129	-.462
-.311	-.440	.178	-.552	.294	-.492	.214	-.439	.201	-.403	.209	-.368
-.023	-.310	.286	-.511	.404	-.284	.295	-.323	.294	-.354	.293	-.346
.133	-.264	.396	-.430	.497	-.286	.407	-.297	.397	-.324	.494	-.230
.272	-.206	.514	-.335	.599	-.245	.502	-.307	.495	-.321	.590	-.233
.416	-.194	.618	-.313	.700	-.276	.601	-.314	.594	-.306	.693	-.247
.565	-.153	.733	-.208	.864	-.223	.698	-.341	.693	-.328	.777	-.254
.713	-.153	.835	-.158	.926	-.123	.863	-.268	.784	-.319	.861	-.199
.854	-.178	.919	-.087			.923	-.156	.856	-.271	.918	-.140
.980	-.159	.987	-.006			.977	-.041	.926	-.141	.972	-.006
1.074	-.121							.977	-.026		
					LOWER SURFACE						
-.660	.047	-.022	.312	.024	.203	.074	-.043	.019	.168	.020	.173
-.616	.025	.038	.084	.075	-.020	.130	-.060	.066	-.054	.076	-.137
-.462	-.039	.101	-.022	.297	-.108	.298	-.096	.136	-.085	.136	-.184
-.329	-.077	.165	-.068	.400	-.097	.397	-.101	.214	-.097	.221	-.167
-.172	-.102	.398	-.110	.604	-.071	.501	-.089	.292	-.081	.295	-.165
-.030	-.148	.737	.124	.785	.191	.603	.003	.403	-.078	.396	-.103
.128	-.181			.967	.209	.784	.170	.489	-.108	.497	-.097
.418	-.082			1.000	.003	.868	.248	.594	-.091	.597	-.033
.564	-.036					.923	.296	.700	.070	.702	.098
.710	.069					.972	.193	.786	.185	.786	.183
.976	.238							.858	.257	.864	.239
1.072	.217							.919	.280	.912	.241
1.110	.163							.967	.159	.985	.069
CN=	.3993		.4131		.4312		.4147		.3863		.2969
CM=	-.0038		-.0462		-.0604		-.0818		-.0788		-.0647

$\alpha = 3.93^{\circ}$; $C_L = 0.443$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.110	-.021	-.766	.023	-.934	.025	-.911	.022	-.820	.018	-.702
-.567	-.272	.035	-.832	.068	-1.069	.079	-.984	.075	-.982	.077	-.883
-.452	-.373	.105	-.601	.209	-.954	.133	-1.006	.129	-.881	.129	-.841
-.311	-.446	.178	-.577	.294	-.662	.214	-.660	.201	-.458	.209	-.379
-.023	-.327	.286	-.561	.404	-.392	.295	-.419	.294	-.344	.293	-.348
.133	-.269	.396	-.502	.497	-.277	.407	-.297	.397	-.330	.494	-.228
.272	-.221	.514	-.410	.599	-.234	.502	-.281	.495	-.334	.590	-.229
.416	-.205	.618	-.348	.700	-.262	.601	-.308	.594	-.327	.693	-.250
.565	-.189	.733	-.214	.864	-.213	.698	-.334	.693	-.331	.777	-.261
.713	-.177	.835	-.146	.926	-.119	.863	-.280	.784	-.330	.861	-.207
.854	-.209	.919	-.070			.923	-.169	.856	-.265	.918	-.154
.980	-.181	.987	-.009			.977	-.048	.926	-.141	.972	-.013
1.074	-.134							.977	-.028		
					LOWER SURFACE						
-.660	.048	-.022	.343	.024	.263	.074	.048	.019	.208	.020	.220
-.616	.047	.038	.124	.075	.023	.130	-.015	.066	-.025	.076	-.051
-.462	-.030	.101	.013	.297	-.084	.298	-.062	.136	-.045	.136	-.129
-.329	-.062	.185	-.034	.400	-.078	.397	-.083	.214	-.057	.221	-.128
-.172	-.068	.398	-.073	.604	-.064	.501	-.077	.292	-.058	.295	-.126
-.030	-.139	.737	.131	.785	.196	.603	.017	.403	-.052	.396	-.099
.128	-.165			.967	.211	.784	.168	.489	-.090	.497	-.087
.418	-.063			1.000	-.001	.868	.250	.594	-.077	.597	-.024
.564	-.030					.923	.305	.700	.077	.702	.100
.710	.078					.972	.201	.786	.186	.786	.180
.976	.244							.858	.263	.864	.232
1.072	.218							.919	.283	.912	.235
1.110	.164							.967	.157	.985	.063
CN=	.4528		.4824		.5137		.4845		.4413		.3564
CM=	-.0031		-.0477		-.0574		-.0788		-.0767		-.0584

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90. Continued.

$\alpha = 4.91^{\circ}$; $C_L = 0.562$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.126	-.021	-.995	.023	-1.042	.025	-1.019	.022	-.969	.018	-.890
-.567	-.309	.035	-1.082	.068	-1.184	.079	-1.102	.075	-1.139	.077	-1.052
-.452	-.372	.105	-.669	.209	-1.049	.133	-1.095	.129	-1.028	.129	-1.055
-.311	-.479	.178	-.623	.294	-.973	.214	-1.058	.201	-1.047	.209	-.931
-.023	-.349	.286	-.608	.404	-.618	.295	-.837	.294	-.626	.293	-.419
.133	-.282	.396	-.576	.497	-.388	.407	-.479	.397	-.428	.494	-.145
.272	-.252	.514	-.481	.599	-.249	.502	-.409	.495	-.317	.590	-.166
.416	-.237	.618	-.452	.700	-.222	.601	-.251	.594	-.277	.693	-.216
.565	-.211	.733	-.253	.864	-.173	.698	-.257	.693	-.273	.777	-.252
.713	-.215	.835	-.132	.926	-.089	.863	-.249	.784	-.255	.861	-.205
.854	-.260	.919	-.063			.923	-.160	.856	-.210	.918	-.158
.980	-.221	.987	-.025			.977	-.038	.926	-.114	.972	-.030
1.074	-.154							.977	-.033		
					LOWER SURFACE						
-.660	.050	-.022	.392	.024	.333	.074	.087	.019	.301	.020	.312
-.616	.055	.038	.166	.075	.096	.130	.025	.066	.069	.076	.033
-.462	-.000	.101	.082	.297	-.044	.298	-.029	.136	.008	.136	-.057
-.329	-.039	.185	.019	.400	-.049	.397	-.056	.214	-.011	.221	-.086
-.172	-.080	.398	-.043	.604	-.049	.501	-.048	.292	-.019	.295	-.091
-.030	-.104	.737	.138	.785	.201	.603	.042	.403	-.017	.396	-.081
.128	-.131			.967	.211	.784	.174	.489	-.058	.497	-.081
.418	-.040			1.000	-.003	.868	.261	.594	-.060	.597	-.019
.564	.002					.923	.318	.700	.086	.702	.056
.710	.098					.972	.218	.786	.192	.786	.176
.976	.259							.858	.264	.864	.229
1.072	.224							.919	.285	.912	.232
1.110	.161							.967	.158	.985	.061
CN=	.5420		.5862		.6205		.6114		.5594		.4452
CM=	-.0049		-.0498		-.0547		-.0766		-.0620		-.0428

$\alpha = 5.90^{\circ}$; $C_L = 0.675$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.154	-.021	-1.129	.023	-1.134	.025	-1.122	.022	-1.086	.018	-1.019
-.567	-.343	.035	-1.281	.068	-1.258	.079	-1.190	.075	-1.223	.077	-1.157
-.452	-.403	.105	-.827	.209	-1.178	.133	-1.163	.129	-1.147	.129	-.984
-.311	-.496	.178	-.711	.294	-1.105	.214	-1.134	.201	-1.142	.209	-.595
-.023	-.429	.286	-.643	.404	-.810	.295	-1.076	.294	-.681	.293	-.553
.133	-.299	.396	-.614	.497	-.611	.407	-.717	.397	-.605	.494	-.327
.272	-.275	.514	-.533	.599	-.396	.502	-.595	.495	-.551	.590	-.239
.416	-.259	.618	-.508	.700	-.210	.601	-.485	.594	-.465	.693	-.192
.565	-.229	.733	-.353	.864	-.109	.698	-.250	.693	-.407	.777	-.166
.713	-.244	.835	-.143	.926	-.072	.863	-.181	.784	-.280	.861	-.125
.854	-.323	.919	-.068			.923	-.131	.856	-.191	.918	-.109
.980	-.298	.987	-.031			.977	-.033	.926	-.114	.972	-.087
1.074	-.188							.977	-.061		
					LOWER SURFACE						
-.660	.052	-.022	.420	.024	.384	.074	.145	.019	.369	.020	.370
-.616	.072	.038	.233	.075	.161	.130	.080	.066	.138	.076	.090
-.462	.016	.101	.137	.297	-.007	.298	.014	.136	.069	.136	-.000
-.329	-.006	.185	.066	.400	-.018	.397	-.021	.214	.028	.221	-.047
-.172	-.043	.398	-.011	.604	-.034	.501	-.024	.292	.008	.295	-.072
-.030	-.073	.737	.148	.785	.203	.603	.052	.403	.002	.396	-.075
.128	-.104			.967	.212	.784	.179	.489	-.041	.497	-.084
.418	-.009			1.000	-.011	.868	.272	.594	-.041	.597	-.035
.564	.022					.923	.329	.700	.086	.702	.084
.710	.123					.972	.223	.786	.188	.786	.155
.976	.275							.858	.259	.864	.210
1.072	.228							.919	.279	.912	.203
1.110	.168							.967	.137	.985	.042
CN=	.6524		.6890		.7302		.7379		.6847		.4708
CM=	-.0051		-.0551		-.0600		-.0864		-.0784		-.0380

[redacted]

TABLE IV – *Continued*

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(d) M = 0.90. Concluded.

$\alpha = 6.94^\circ$; $C_L = 0.776$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.199	-.021	-1.251	.023	-1.225	.025	-1.209	.022	-1.168	.018	-.592
-.567	-.398	.035	-1.358	.068	-1.328	.079	-1.270	.075	-1.295	.077	-.589
-.452	-.460	.105	-.926	.209	-1.226	.133	-1.226	.129	-1.226	.129	-.547
-.311	-.519	.178	-.883	.294	-1.169	.214	-1.202	.201	-1.197	.209	-.503
-.023	-.475	.286	-.750	.404	-.900	.295	-1.150	.294	-.695	.293	-.479
.133	-.326	.396	-.677	.497	-.701	.407	-.786	.397	-.658	.494	-.400
.272	-.284	.514	-.585	.599	-.631	.502	-.710	.495	-.635	.590	-.363
.416	-.282	.618	-.548	.700	-.467	.601	-.593	.594	-.568	.693	-.323
.565	-.254	.733	-.429	.864	-.122	.698	-.342	.693	-.475	.777	-.277
.713	-.274	.835	-.192	.926	-.061	.863	-.143	.784	-.398	.861	-.251
.854	-.351	.919	-.083			.923	-.125	.856	-.306	.918	-.246
.980	-.374	.987	-.034			.977	-.069	.926	-.269	.972	-.226
1.074	-.242							.977	-.225		
				LOWER SURFACE							
-.660	.052	-.022	.443	.024	.444	.074	.202	.019	.409	.020	.415
-.616	.086	.038	.281	.075	.221	.130	.125	.066	.183	.076	.142
-.462	.050	.101	.191	.297	.047	.298	.030	.136	.097	.136	.032
-.329	.021	.185	.118	.400	.007	.397	-.002	.214	.058	.221	-.015
-.172	-.017	.398	.029	.604	-.022	.501	-.028	.292	.025	.295	-.052
-.030	-.040	.737	.157	.785	.207	.603	.048	.403	.018	.396	-.078
.128	-.071			.967	.201	.784	.149	.489	-.026	.497	-.106
.418	.029			1.000	-.019	.868	.260	.594	-.036	.597	-.073
.564	.053					.923	.326	.700	.087	.702	.030
.710	.143					.972	.202	.786	.179	.786	.103
.976	.265							.858	.247	.864	.157
1.072	.237							.919	.258	.912	.157
1.110	.163							.967	.095	.985	-.133
CN=	.7638		.7990		.8541		.8097		.7776		.4219
CM=	.0004		-.0639		-.0846		-.0907		-.1037		-.0664

$\alpha = 7.97^\circ$; $C_L = 0.834$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.234	-.021	-1.312	.023	-1.300	.025	-1.275	.022	-1.234	.018	-.611
-.567	-.476	.035	-1.445	.068	-1.391	.079	-1.330	.075	-1.350	.077	-.602
-.452	-.484	.105	-1.056	.209	-1.270	.133	-1.286	.129	-1.260	.129	-.558
-.311	-.551	.178	-1.035	.294	-.939	.214	-1.008	.201	-1.060	.209	-.531
-.023	-.493	.286	-.979	.404	-.774	.295	-.843	.294	-.795	.293	-.456
.133	-.360	.396	-.819	.497	-.757	.407	-.668	.397	-.756	.494	-.429
.272	-.322	.514	-.708	.599	-.703	.502	-.390	.495	-.702	.590	-.391
.416	-.312	.618	-.589	.700	-.618	.601	-.289	.594	-.632	.693	-.359
.565	-.287	.733	-.446	.864	-.455	.698	-.280	.693	-.548	.777	-.318
.713	-.301	.835	-.217	.926	-.408	.863	-.262	.784	-.457	.861	-.296
.854	-.379	.919	-.092			.923	-.248	.856	-.361	.918	-.286
.980	-.419	.987	-.034			.977	-.252	.926	-.277	.972	-.269
1.074	-.250							.977	-.207		
				LOWER SURFACE							
-.660	.042	-.022	.449	.024	.481	.074	.240	.019	.429	.020	.431
-.616	.106	.038	.337	.075	.278	.130	.152	.066	.193	.076	.165
-.462	.083	.101	.241	.297	.076	.298	.039	.136	.120	.136	.066
-.329	.048	.185	.165	.400	.036	.397	-.008	.214	.051	.221	.005
-.172	.013	.398	.061	.604	-.018	.501	-.050	.292	.028	.295	-.039
-.030	-.010	.737	.173	.785	.182	.603	.014	.403	-.005	.396	-.062
.128	-.031			.967	.105	.784	.099	.489	-.036	.497	-.097
.418	.054			1.000	-.278	.868	.216	.594	-.054	.597	-.073
.564	.084					.923	.270	.700	.085	.702	.037
.710	.164					.972	.109	.786	.187	.786	.102
.976	.299							.858	.262	.864	.147
1.072	.241							.919	.273	.912	.147
1.110	.175							.967	.108	.985	-.149
CN=	.8673		.9272		.9436		.7066		.8234		.4586
CM=	.0100		-.0736		-.1293		-.0697		-.1164		-.0749

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95.

$\alpha = -4.97^{\circ}$; $C_L = -0.546$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	.047	-.021	.362	.023	.358	.025	.349	.022	.432	.018	.469
-.567	.002	.035	.192	.068	.210	.079	.199	.075	.233	.077	.238
-.452	-.113	.105	.116	.209	.074	.133	.143	.129	.166	.129	.159
-.311	-.202	.178	.068	.294	-.012	.214	.097	.201	.104	.209	.095
-.023	-.091	.286	.026	.404	.015	.295	.073	.294	.058	.293	.027
.133	-.019	.396	-.002	.497	-.016	.407	.018	.397	.018	.494	-.108
.272	.020	.514	-.003	.599	-.054	.502	-.038	.495	-.047	.590	-.178
.416	.053	.618	-.035	.700	-.100	.601	-.092	.594	-.099	.693	-.267
.565	.082	.733	-.062	.864	-.209	.698	-.186	.693	-.177	.777	-.384
.713	.072	.835	-.104	.926	-.255	.863	-.401	.784	-.257	.861	-.480
.854	.025	.919	-.085			.923	-.337	.856	-.362	.918	-.636
.980	-.019	.987	.034			.977	-.124	.926	-.384	.972	-.327
1.074	-.080							.977	-.394		
					LOWER SURFACE						
-.660	-.056	-.022	-.655	.024	-1.008	.074	-1.198	.019	-.983	.020	-.424
-.616	-.150	.038	-.921	.075	-1.031	.130	-1.135	.066	-.889	.076	-.401
-.462	-.232	.101	-.775	.297	-1.008	.298	-.841	.136	-.839	.136	-.397
-.329	-.249	.185	-.633	.400	-.975	.397	-.544	.214	-.783	.221	-.372
-.172	-.251	.398	-.629	.604	-.385	.501	-.616	.292	-.728	.295	-.360
-.030	-.294	.737	-.226	.785	-.004	.603	-.437	.403	-.674	.396	-.323
.128	-.322			.967	.023	.784	-.062	.489	-.626	.497	-.290
.418	-.300			1.000	-.049	.868	-.025	.594	-.403	.597	-.272
.564	-.352					.923	-.058	.700	-.478	.702	-.260
.710	-.323					.972	-.058	.786	-.408	.786	-.253
.976	.024							.858	-.379	.864	-.217
1.072	.090							.919	-.371	.912	-.231
1.110	.076							.967	-.347	.985	-.227
CN=	-.3608		-.5672		-.5505		-.5077		-.5376		-.1872
CM=	-.0084		.0148		-.0273		-.0420		.0181		-.0456

$\alpha = -3.97^{\circ}$; $C_L = -0.471$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	.041	-.021	.296	.023	.286	.025	.281	.022	.409	.018	.452
-.567	-.011	.035	.127	.068	.133	.079	.155	.075	.207	.077	.208
-.452	-.129	.105	.077	.209	.015	.133	.109	.129	.136	.129	.133
-.311	-.247	.178	.016	.294	-.055	.214	.054	.201	.085	.209	.065
-.023	-.096	.286	-.015	.404	-.027	.295	.041	.294	.056	.293	.006
.133	-.044	.396	-.043	.497	-.058	.407	-.010	.397	.023	.494	-.130
.272	-.001	.514	-.038	.599	-.080	.502	-.048	.495	-.033	.590	-.204
.416	.031	.618	-.062	.700	-.120	.601	-.098	.594	-.083	.693	-.311
.565	.060	.733	-.085	.864	-.168	.698	-.170	.693	-.161	.777	-.411
.713	.057	.835	-.121	.926	-.166	.863	-.215	.784	-.249	.861	-.521
.854	.004	.919	-.090			.923	-.167	.856	-.384	.918	-.645
.980	-.042	.987	.023			.977	-.025	.926	-.514	.972	-.262
1.074	-.084							.977	-.526		
					LOWER SURFACE						
-.660	-.043	-.022	-.509	.024	-.912	.074	-1.145	.019	-.979	.020	-.410
-.616	-.116	.038	-.756	.075	-.932	.130	-1.116	.066	-.979	.076	-.391
-.462	-.206	.101	-.596	.297	-.933	.298	-1.027	.136	-.906	.136	-.362
-.329	-.246	.185	-.511	.400	-.896	.397	-.726	.214	-.769	.221	-.339
-.172	-.248	.398	-.594	.604	-.193	.501	-.318	.292	-.698	.295	-.324
-.030	-.317	.737	-.161	.785	-.048	.603	-.182	.403	-.629	.396	-.285
.128	-.343			.967	.063	.784	.216	.489	-.587	.497	-.268
.418	-.285			1.000	-.039	.868	.193	.594	-.529	.597	-.245
.564	-.327					.923	-.005	.700	-.435	.702	-.223
.710	-.279					.972	-.008	.786	-.344	.786	-.237
.976	.050							.858	-.278	.864	-.220
1.072	.104							.919	-.224	.912	-.221
1.110	.092							.967	-.213	.985	-.219
CN=	-.3024		-.4384		-.4527		-.4378		-.5009		-.1414
CM=	-.0121		.0037		-.0313		-.0654		-.0082		-.0536

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(e) M = 0.95. Continued.

$\alpha = -2.99^0$; $C_L = -0.368$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	.033	-.021	.258	.023	.194	.025	.221	.022	.341	.018	.429
	-.567	-.042	.035	.044	.068	.047	.079	.078	.075	.153	.077	.159
	-.452	-.164	.105	-.007	.209	-.042	.133	.055	.129	.103	.129	.099
	-.311	-.267	.178	-.054	.294	-.086	.214	.014	.201	.040	.209	.043
	-.023	-.205	.286	-.077	.404	-.071	.295	.015	.294	.019	.293	-.021
	.133	-.064	.396	-.090	.497	-.092	.407	-.036	.397	-.011	.494	-.130
	.272	-.035	.514	-.077	.599	-.111	.502	-.071	.495	-.058	.590	-.199
	.416	-.001	.618	-.099	.700	-.144	.601	-.114	.594	-.108	.693	-.275
	.565	.028	.733	-.118	.864	-.171	.698	-.183	.693	-.186	.777	-.384
	.713	.024	.835	-.142	.926	-.119	.863	-.252	.784	-.258	.861	-.443
	.854	-.023	.919	-.105			.923	-.164	.856	-.321	.918	-.587
	.980	-.060	.987	.015			.977	-.017	.926	-.333	.972	-.299
	1.074	-.093							.977	-.182		
						LOWER SURFACE						
	-.660	-.011	-.022	-.291	.024	-.809	.074	-1.072	.019	-1.001	.020	-.484
	-.616	-.100	.038	-.525	.075	-.798	.130	-1.030	.066	-.984	.076	-.472
	-.462	-.183	.101	-.459	.297	-.811	.298	-.982	.136	-.978	.136	-.394
	-.329	-.219	.185	-.503	.400	-.780	.397	-.845	.214	-.793	.221	-.367
	-.172	-.243	.398	-.537	.604	-.230	.501	-.218	.292	-.702	.295	-.347
	-.030	-.324	.737	-.105	.785	.095	.603	-.048	.403	-.578	.396	-.308
	.128	-.376			.967	.121	.784	.189	.489	-.549	.497	-.273
	.418	-.277			1.000	.062	.868	.208	.594	-.508	.597	-.256
	.564	-.297					.923	.186	.700	-.406	.702	-.230
	.710	-.231					.972	.108	.786	-.282	.786	-.218
	.976	.079							.858	-.180	.864	-.209
	1.072	.128							.919	-.046	.912	-.216
	1.110	.109							.967	.119	.985	-.220
CN=	-.1996		-.3156		-.3350		-.3548		-.4612		-.1595	
CM=	-.0101		-.0075		-.0454		-.0836		-.0191		-.0480	

$\alpha = -2.12^o$; $C_L = -0.270$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	.030	-.021	.196	.023	.133	.025	.144	.022	.297	.018	.381
	-.567	-.040	.035	-.005	.068	-.022	.079	.025	.075	.102	.077	.120
	-.452	-.178	.105	-.063	.209	-.095	.133	-.002	.129	.059	.129	.051
	-.311	-.275	.178	-.083	.294	-.139	.214	-.032	.201	.009	.209	-.001
	-.023	-.350	.286	-.128	.404	-.097	.295	-.030	.294	-.017	.293	-.050
	.133	-.061	.396	-.125	.497	-.119	.407	-.066	.397	-.035	.494	-.134
	.272	-.040	.514	-.105	.599	-.131	.502	-.094	.495	-.092	.590	-.194
	.416	-.016	.618	-.126	.700	-.162	.601	-.134	.594	-.137	.693	-.192
	.565	.008	.733	-.129	.864	-.181	.698	-.208	.693	-.206	.777	-.237
	.713	.006	.835	-.149	.926	-.102	.863	-.257	.784	-.274	.861	-.269
	.854	-.031	.919	-.098			.923	-.168	.856	-.285	.918	-.319
	.980	-.067	.987	.021			.977	-.017	.926	-.198	.972	-.252
	1.074	-.079							.977	-.014		
						LOWER SURFACE						
	-.660	-.000	-.022	-.147	.024	-.685	.074	-1.005	.019	-.924	.020	-.614
	-.616	-.079	.038	-.423	.075	-.698	.130	-.980	.066	-.953	.076	-.568
	-.462	-.173	.101	-.440	.297	-.729	.298	-.930	.136	-.964	.136	-.591
	-.329	-.208	.185	-.453	.400	-.613	.397	-.586	.214	-.836	.221	-.425
	-.172	-.238	.398	-.507	.604	-.183	.501	-.128	.292	-.659	.295	-.422
	-.030	-.327	.737	-.024	.785	.132	.603	.005	.403	-.546	.396	-.326
	.128	-.390			.967	.183	.784	.161	.489	-.401	.497	-.290
	.418	-.258			1.000	.051	.868	.219	.594	-.337	.597	-.270
	.564	-.285					.923	.216	.700	-.155	.702	-.235
	.710	-.195					.972	.158	.786	.065	.786	-.183
	.976	.112							.858	.186	.864	-.169
	1.072	.146							.919	.228	.912	-.134
	1.110	.122							.967	.171	.985	-.065
CN=	-.1434		-.2181		-.2238		-.2607		-.3184		-.2182	
CM=	-.0095		-.0228		-.0590		-.0944		-.0702		-.0321	

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(e) M = 0.95. Continued.

$\alpha = -1.19^o$; $C_L = -0.149$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.009	-.021	.110	.023	.015	.025	.071	.022	.215	.018	.324
-.567	-.068	.035	-.103	.068	-.126	.079	-.052	.075	.021	.077	.058
-.452	-.208	.105	-.136	.209	-.168	.133	-.062	.129	-.017	.129	.005
-.311	-.305	.178	-.151	.294	-.204	.214	-.080	.201	-.046	.209	-.051
-.023	-.384	.286	-.168	.404	-.145	.295	-.083	.294	-.060	.293	-.094
.133	-.081	.396	-.180	.497	-.160	.407	-.103	.397	-.076	.494	-.199
.272	-.063	.514	-.151	.599	-.158	.502	-.140	.495	-.134	.590	-.245
.416	-.051	.618	-.165	.700	-.211	.601	-.163	.594	-.157	.693	-.212
.565	-.020	.733	-.159	.864	-.208	.698	-.232	.693	-.223	.777	-.181
.713	-.019	.835	-.166	.926	-.121	.863	-.282	.784	-.315	.861	-.169
.854	-.064	.919	-.103			.923	-.174	.856	-.331	.918	-.117
.980	-.093	.987	.015			.977	-.013	.926	-.191	.972	.011
1.074	-.096							.977	-.015		

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
-.660	.010	-.022	-.055	.024	-.535	.074	-.900	.019	-.784	.020	-.845
-.616	-.050	.038	-.310	.075	-.583	.130	-.871	.066	-.908	.076	-.748
-.462	-.140	.101	-.373	.297	-.592	.298	-.371	.136	-.882	.136	-.742
-.329	-.180	.185	-.406	.400	-.439	.397	-.221	.214	-.668	.221	-.598
-.172	-.225	.398	-.443	.604	-.179	.501	-.193	.292	-.447	.295	-.543
-.030	-.317	.737	.041	.785	.152	.603	-.013	.403	-.330	.396	-.423
.128	-.374			.967	.226	.784	.155	.489	-.269	.497	-.273
.418	-.233			1.000	.032	.868	.205	.594	-.155	.597	-.146
.564	-.254					.923	.234	.700	-.019	.702	-.016
.710	-.124					.972	.166	.786	.073	.786	.062
.976	.145							.858	.155	.864	.117
1.072	.170							.919	.192	.912	.158
1.110	.141							.967	.126	.985	.088

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	-.0408	-.0988	-.0896	-.0974	-.1704	-.1874
CM=	-.0130	-.0388	-.0714	-.1013	-.0869	-.0684

$\alpha = -0.03^o$; $C_L = -0.003$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.001	-.021	.022	.023	-.150	.025	-.093	.022	.099	.018	.205
-.567	-.087	.035	-.207	.068	-.268	.079	-.182	.075	-.100	.077	-.051
-.452	-.226	.105	-.217	.209	-.238	.133	-.173	.129	-.112	.129	-.094
-.311	-.329	.178	-.237	.294	-.228	.214	-.155	.201	-.140	.209	-.112
-.023	-.420	.286	-.235	.404	-.174	.295	-.147	.294	-.126	.293	-.143
.133	-.216	.396	-.246	.497	-.225	.407	-.147	.397	-.120	.494	-.227
.272	-.068	.514	-.188	.599	-.215	.502	-.184	.495	-.189	.590	-.291
.416	-.064	.618	-.195	.700	-.228	.601	-.223	.594	-.236	.693	-.307
.565	-.043	.733	-.171	.864	-.235	.698	-.284	.693	-.273	.777	-.126
.713	-.039	.835	-.164	.926	-.131	.863	-.293	.784	-.309	.861	-.145
.854	-.095	.919	-.106			.923	-.155	.856	-.308	.918	-.106
.980	-.113	.987	.014			.977	-.002	.926	-.196	.972	.033
1.074	-.100							.977	-.014		

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
-.660	.025	-.022	.078	.024	-.333	.074	-.691	.019	-.573	.020	-.601
-.616	-.033	.038	-.184	.075	-.501	.130	-.619	.066	-.766	.076	-.783
-.462	-.118	.101	-.286	.297	-.437	.298	-.256	.136	-.708	.136	-.736
-.329	-.160	.185	-.319	.400	-.158	.397	-.248	.214	-.361	.221	-.571
-.172	-.213	.398	-.369	.604	-.184	.501	-.262	.292	-.239	.295	-.388
-.030	-.291	.737	.088	.785	.163	.603	-.020	.403	-.202	.396	-.257
.128	-.337			.967	.221	.784	.126	.489	-.194	.497	-.080
.418	-.200			1.000	.022	.868	.192	.594	-.190	.597	-.001
.564	-.193					.923	.197	.700	.020	.702	.113
.710	-.048					.972	.149	.786	.127	.786	.165
.976	.180							.858	.186	.864	.192
1.072	.190							.919	.219	.912	.199
1.110	.151							.967	.155	.985	.110

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.0757	.0279	.0522	.0057	-.0147	-.0420
CM=	-.0178	-.0467	-.0784	-.0905	-.0940	-.0930

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Continued.

$\alpha = 1.02^{\circ}$; $C_L = 0.122$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.023	-.021	-.104	.023	-.360	.025	-.297	.022	-.087	.018	.054
-.567	-.116	.035	-.298	.068	-.473	.079	-.345	.075	-.231	.077	-.189
-.452	-.249	.105	-.295	.209	-.361	.133	-.289	.129	-.223	.129	-.190
-.311	-.352	.178	-.311	.294	-.364	.214	-.214	.201	-.219	.209	-.202
-.023	-.443	.286	-.291	.404	-.243	.295	-.188	.294	-.199	.293	-.217
.133	-.314	.396	-.279	.497	-.254	.407	-.216	.397	-.187	.494	-.252
.272	-.081	.514	-.256	.599	-.192	.502	-.241	.495	-.226	.590	-.307
.416	-.080	.618	-.250	.700	-.246	.601	-.266	.594	-.261	.693	-.283
.565	-.059	.733	-.240	.864	-.277	.698	-.345	.693	-.322	.777	-.158
.713	-.068	.835	-.152	.926	-.150	.863	-.284	.784	-.345	.861	-.168
.854	-.128	.919	-.088			.923	-.140	.856	-.380	.918	-.127
.980	-.152	.987	.013			.977	-.010	.926	-.186	.972	.021
1.074	-.122							.977	-.017		
					LOWER SURFACE						
-.660	.038	-.022	.170	.024	-.109	.074	-.353	.019	-.347	.020	-.302
-.616	-.012	.038	-.090	.075	-.321	.130	-.231	.066	-.502	.076	-.498
-.462	-.100	.101	-.186	.297	-.269	.298	-.286	.136	-.386	.136	-.474
-.329	-.140	.185	-.237	.400	-.161	.397	-.235	.214	-.192	.221	-.428
-.172	-.197	.398	-.238	.604	-.083	.501	-.165	.292	-.242	.295	-.320
-.030	-.253	.737	.097	.785	.178	.603	-.004	.403	-.199	.396	-.272
.128	-.321			.967	.230	.784	.145	.489	-.174	.497	-.074
.418	-.156			1.000	.025	.868	.195	.594	-.163	.597	-.008
.564	-.141					.923	.211	.700	.043	.702	.123
.710	-.006					.972	.162	.786	.153	.786	.195
.976	.205							.858	.218	.864	.230
1.072	.203							.919	.246	.912	.234
1.110	.161							.967	.165	.985	.111
CN=	.1809		.1551		.2109		.1566		.1225		.0798
CM=	-.0210		-.0525		-.0832		-.0869		-.0971		-.0894

$\alpha = 1.99^{\circ}$; $C_L = 0.238$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.034	-.021	-.454	.023	-.496	.025	-.484	.022	-.274	.018	-.132
-.567	-.136	.035	-.414	.068	-.635	.079	-.577	.075	-.359	.077	-.365
-.452	-.278	.105	-.369	.209	-.493	.133	-.527	.129	-.309	.129	-.328
-.311	-.382	.178	-.388	.294	-.450	.214	-.470	.201	-.286	.209	-.288
-.023	-.468	.286	-.344	.404	-.312	.295	-.243	.294	-.279	.293	-.283
.133	-.361	.396	-.351	.497	-.322	.407	-.283	.397	-.256	.494	-.328
.272	-.169	.514	-.306	.599	-.380	.502	-.152	.495	-.281	.590	-.370
.416	-.066	.618	-.317	.700	-.269	.601	-.210	.594	-.339	.693	-.132
.565	-.078	.733	-.302	.864	-.210	.698	-.310	.693	-.423	.777	-.170
.713	-.086	.835	-.235	.926	-.123	.863	-.386	.784	-.494	.861	-.186
.854	-.153	.919	-.077			.923	-.142	.856	-.259	.918	-.135
.980	-.201	.987	.012			.977	-.036	.926	-.160	.972	.020
1.074	-.150							.977	-.025		
					LOWER SURFACE						
-.660	.044	-.022	.241	.024	.054	.074	-.190	.019	-.078	.020	-.092
-.616	.004	.038	-.016	.075	-.202	.130	-.179	.066	-.320	.076	-.360
-.462	-.068	.101	-.106	.297	-.211	.298	-.221	.136	-.239	.136	-.374
-.329	-.116	.185	-.182	.400	-.145	.397	-.162	.214	-.211	.221	-.315
-.172	-.176	.398	-.176	.604	-.074	.501	-.135	.292	-.221	.295	-.295
-.030	-.222	.737	.107	.785	.186	.603	-.001	.403	-.106	.396	-.194
.128	-.311			.967	.226	.784	.157	.489	-.151	.497	-.108
.418	-.136			1.000	.023	.868	.220	.594	-.165	.597	-.019
.564	-.101					.923	.261	.700	.051	.702	.119
.710	.024					.972	.190	.786	.169	.786	.207
.976	.213							.858	.245	.864	.253
1.072	.216							.919	.271	.912	.254
1.110	.153							.967	.164	.985	.100
CN=	.2724		.2778		.3194		.2716		.2389		.1727
CM=	-.0177		-.0591		-.0805		-.0866		-.1021		-.0818

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Continued.

$\alpha = 2.48^{\circ}$; $C_L = 0.295$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.043	-.021	-.591	.023	-.579	.025	-.612	.022	-.461	.018	-.260
-.567	-.157	.035	-.761	.068	-.736	.079	-.696	.075	-.577	.077	-.397
-.452	-.286	.105	-.224	.209	-.578	.133	-.673	.129	-.532	.129	-.380
-.311	-.394	.178	-.353	.294	-.532	.214	-.602	.201	-.405	.209	-.334
-.023	-.482	.286	-.385	.404	-.385	.295	-.322	.294	-.144	.293	-.327
.133	-.386	.396	-.379	.497	-.347	.407	-.331	.397	-.205	.494	-.353
.272	-.148	.514	-.335	.599	-.409	.502	-.326	.495	-.243	.590	-.374
.416	-.078	.618	-.337	.700	-.388	.601	-.164	.594	-.332	.693	-.129
.565	-.083	.733	-.332	.864	-.172	.698	-.262	.693	-.397	.777	-.180
.713	-.108	.835	-.282	.926	-.091	.863	-.388	.784	-.483	.861	-.190
.854	-.175	.919	-.092			.923	-.186	.856	-.459	.918	-.137
.980	-.226	.987	-.008			.977	-.034	.926	-.114	.972	.015
1.074	-.192							.977	-.033		
					LOWER SURFACE						
-.660	.049	-.022	.284	.024	.082	.074	-.143	.019	-.030	.020	-.004
-.616	.017	.038	.027	.075	-.135	.130	-.161	.066	-.168	.076	-.260
-.462	-.062	.101	-.068	.297	-.178	.298	-.143	.136	-.186	.136	-.290
-.329	-.101	.185	-.128	.400	-.134	.397	-.145	.214	-.153	.221	-.260
-.172	-.150	.398	-.152	.604	-.079	.501	-.118	.292	-.142	.295	-.260
-.030	-.205	.737	.111	.785	.187	.603	.008	.403	-.084	.396	-.180
.128	-.288			.967	.230	.784	.161	.489	-.137	.497	-.110
.418	-.118			1.000	.027	.868	.236	.594	-.159	.597	-.022
.564	-.080					.923	.276	.700	.059	.702	.117
.710	.036					.972	.198	.786	.177	.786	.200
.976	.220							.858	.245	.864	.246
1.072	.210							.919	.269	.912	.254
1.110	.151							.967	.158	.985	.101
CN=	.3246		.3424		.3879		.3511		.3013		.2227
CM=	-.0189		-.0621		-.0815		-.0875		-.0976		-.0783

$\alpha = 2.84^{\circ}$; $C_L = 0.341$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.060	-.021	-.659	.023	-.633	.025	-.654	.022	-.534	.018	-.397
-.567	-.188	.035	-.767	.068	-.790	.079	-.727	.075	-.711	.077	-.470
-.452	-.300	.105	-.279	.209	-.556	.133	-.737	.129	-.652	.129	-.496
-.311	-.404	.178	-.395	.294	-.633	.214	-.655	.201	-.610	.209	-.283
-.023	-.489	.286	-.396	.404	-.434	.295	-.505	.294	-.085	.293	-.276
.133	-.380	.396	-.390	.497	-.368	.407	-.343	.397	-.152	.494	-.330
.272	-.262	.514	-.346	.599	-.414	.502	-.378	.495	-.175	.590	-.368
.416	-.076	.618	-.353	.700	-.437	.601	-.402	.594	-.270	.693	-.397
.565	-.073	.733	-.350	.864	-.140	.698	-.179	.693	-.373	.777	-.112
.713	-.108	.835	-.308	.926	-.061	.863	-.330	.784	-.439	.861	-.135
.854	-.178	.919	-.117			.923	-.181	.856	-.491	.918	-.100
.980	-.236	.987	-.017			.977	-.019	.926	-.157	.972	.020
1.074	-.220							.977	-.042		
					LOWER SURFACE						
-.660	.048	-.022	.293	.024	.111	.074	-.107	.019	.085	.020	.076
-.616	.024	.038	.057	.075	-.108	.130	-.152	.066	-.133	.076	-.193
-.462	-.049	.101	-.048	.297	-.159	.298	-.137	.136	-.144	.136	-.271
-.329	-.101	.185	-.096	.400	-.133	.397	-.147	.214	-.145	.221	-.233
-.172	-.165	.398	-.140	.604	-.075	.501	-.115	.292	-.123	.295	-.227
-.030	-.203	.737	.117	.785	.191	.603	.014	.403	-.076	.396	-.174
.128	-.288			.967	.230	.784	.171	.489	-.121	.497	-.125
.418	-.109			1.000	.022	.868	.246	.594	-.124	.597	-.033
.564	-.068					.923	.283	.700	.064	.702	.107
.710	.045					.972	.215	.786	.177	.786	.189
.976	.230							.858	.252	.864	.241
1.072	.216							.919	.273	.912	.240
1.110	.150							.967	.149	.985	.081
CN=	.3608		.3822		.4197		.4023		.3311		.2466
CM=	-.0184		-.0662		-.0806		-.0885		-.0897		-.0734

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(e) M = 0.95. Continued.

$\alpha = 3.42^o$; $C_L = 0.412$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
UPPER SURFACE											
-.660	-.066	-.021	-.770	.023	-.734	.025	-.705	.022	-.583	.018	-.520
-.567	-.211	.035	-.839	.068	-.861	.079	-.814	.075	-.776	.077	-.747
-.452	-.302	.105	-.800	.209	-.728	.133	-.803	.129	-.769	.129	-.694
-.311	-.412	.178	-.225	.294	-.622	.214	-.769	.201	-.752	.209	-.665
-.023	-.499	.286	-.363	.404	-.464	.295	-.693	.294	-.688	.293	-.623
.133	-.391	.396	-.381	.497	-.403	.407	-.407	.397	-.633	.494	-.182
.272	-.261	.514	-.341	.599	-.422	.502	-.371	.495	-.101	.590	-.257
.416	-.092	.618	-.356	.700	-.464	.601	-.419	.594	-.118	.693	-.258
.565	-.078	.733	-.350	.864	-.143	.698	-.245	.693	-.239	.777	-.241
.713	-.107	.835	-.344	.926	-.060	.863	-.232	.784	-.363	.861	-.189
.854	-.182	.919	-.133			.923	-.142	.856	-.413	.918	-.141
.980	-.241	.987	-.027			.977	-.006	.926	-.172	.972	.012
1.074	-.263							.977	.003		
LOWER SURFACE											
-.660	.050	-.022	.315	.024	.200	.074	-.076	.019	.116	.020	.187
-.616	.036	.038	.086	.075	-.053	.130	-.103	.066	-.101	.076	-.105
-.462	-.034	.101	-.014	.297	-.138	.298	-.109	.136	-.124	.136	-.156
-.329	-.087	.185	-.060	.400	-.128	.397	-.122	.214	-.110	.221	-.159
-.172	-.127	.398	-.118	.604	-.078	.501	-.099	.292	-.100	.295	-.174
-.030	-.183	.737	.115	.785	.187	.603	.021	.403	-.058	.396	-.131
.128	-.254			.967	.216	.784	.175	.489	-.102	.497	-.102
.418	-.101			1.000	-.018	.868	.256	.594	-.099	.597	-.011
.564	-.053					.923	.301	.700	.081	.702	.114
.710	.055					.972	.221	.786	.194	.786	.191
.976	.232							.858	.267	.864	.246
1.072	.205							.919	.291	.912	.240
1.110	.142							.967	.179	.985	.091
CN=	.4050		.4320		.4734		.4641		.4380		.3730
CM=	-.0129		-.0615		-.0770		-.0855		-.0814		-.0637

$\alpha = 3.94^o$; $C_L = 0.478$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
UPPER SURFACE											
-.660	-.079	-.021	-.853	.023	-.795	.025	-.758	.022	-.662	.018	-.603
-.567	-.230	.035	-.942	.068	-.911	.079	-.860	.075	-.837	.077	-.823
-.452	-.320	.105	-.852	.209	-.802	.133	-.846	.129	-.816	.129	-.823
-.311	-.424	.178	-.311	.294	-.688	.214	-.845	.201	-.817	.209	-.799
-.023	-.506	.286	-.399	.404	-.452	.295	-.778	.294	-.764	.293	-.741
.133	-.399	.396	-.415	.497	-.418	.407	-.679	.397	-.755	.494	-.094
.272	-.324	.514	-.365	.599	-.438	.502	-.411	.495	-.602	.590	-.006
.416	-.115	.618	-.377	.700	-.494	.601	-.429	.594	-.153	.693	-.084
.565	-.075	.733	-.372	.864	-.168	.698	-.501	.693	-.173	.777	-.188
.713	-.112	.835	-.357	.926	-.080	.863	-.190	.784	-.281	.861	-.207
.854	-.193	.919	-.154			.923	-.099	.856	-.280	.918	-.182
.980	-.256	.987	-.038			.977	-.010	.926	-.164	.972	-.013
1.074	-.291							.977	-.001		
LOWER SURFACE											
-.660	.057	-.022	.347	.024	.239	.074	-.029	.019	.176	.020	.225
-.616	.052	.038	.133	.075	.015	.130	-.077	.066	-.059	.076	-.061
-.462	-.021	.101	.026	.297	-.111	.298	-.094	.136	-.086	.136	-.137
-.329	-.070	.185	-.035	.400	-.110	.397	-.121	.214	-.087	.221	-.149
-.172	-.120	.398	-.094	.604	-.074	.501	-.099	.292	-.080	.295	-.142
-.030	-.166	.737	.120	.785	.186	.603	.025	.403	-.049	.396	-.126
.128	-.224			.967	.204	.784	.177	.489	-.088	.497	-.110
.418	-.084			1.000	-.030	.868	.259	.594	-.079	.597	-.015
.564	-.036					.923	.313	.700	.094	.702	.117
.710	.075					.972	.224	.786	.207	.786	.198
.976	.240							.858	.282	.864	.244
1.072	.215							.919	.300	.912	.246
1.110	.139							.967	.196	.985	.110
CN=	.4635		.4939		.5239		.5577		.5237		.3741
CM=	-.0110		-.0639		-.0791		-.1005		-.0850		-.0434

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Continued.

$\alpha = 4.94^{\circ}$; $C_L = 0.599$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.107	-.021	-.946	.023	-.894	.025	-.855	.022	-.786	.018	-.718
-.567	-.257	.035	-1.070	.068	-1.018	.079	-.939	.075	-.922	.077	-.894
-.452	-.347	.105	-1.042	.209	-.918	.133	-.941	.129	-.904	.129	-.902
-.311	-.438	.178	-.795	.294	-.892	.214	-.920	.201	-.894	.209	-.873
-.023	-.518	.286	-.309	.404	-.524	.295	-.873	.294	-.867	.293	-.842
.133	-.427	.396	-.380	.497	-.451	.407	-.834	.397	-.831	.494	-.801
.272	-.329	.514	-.379	.599	-.452	.502	-.820	.495	-.817	.590	-.232
.416	-.130	.618	-.393	.700	-.531	.601	-.533	.594	-.794	.693	-.101
.565	-.110	.733	-.389	.864	-.182	.698	-.510	.693	-.353	.777	-.052
.713	-.134	.835	-.379	.926	-.098	.863	-.161	.784	-.227	.861	-.069
.854	-.218	.919	-.171			.923	-.079	.856	-.129	.918	-.096
.980	-.279	.987	-.067			.977	-.031	.926	-.057	.972	.011
1.074	-.336							.977	-.054		
					LOWER SURFACE						
-.660	.060	-.022	.373	.024	.323	.074	.044	.019	.226	.020	.270
-.616	.064	.038	.186	.075	.083	.130	-.008	.066	.002	.076	-.017
-.462	-.002	.101	.075	.297	-.062	.298	-.064	.136	-.045	.136	-.087
-.329	-.044	.185	.019	.400	-.076	.397	-.081	.214	-.055	.221	-.119
-.172	-.080	.398	-.058	.604	-.066	.501	-.076	.292	-.061	.295	-.131
-.030	-.135	.737	.129	.785	.192	.603	.031	.403	-.037	.396	-.132
.128	-.165			.967	.198	.784	.172	.489	-.085	.497	-.125
.418	-.048			1.000	-.053	.868	.259	.594	-.090	.597	-.051
.564	-.012					.923	.313	.700	.079	.702	.091
.710	.087					.972	.208	.786	.190	.786	.175
.976	.252							.858	.264	.864	.231
1.072	.223							.919	.287	.912	.238
1.110	.139							.967	.158	.985	.121
CN=	.5500		.5930		.6200		.6776		.6529		.5120
CM=	-.0088		-.0612		-.0827		-.1113		-.1009		-.0534

$\alpha = 5.87^{\circ}$; $C_L = 0.689$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.152	-.021	-1.005	.023	-.974	.025	-.941	.022	-.861	.018	-.805
-.567	-.294	.035	-1.151	.068	-1.084	.079	-1.002	.075	-.995	.077	-.976
-.452	-.368	.105	-1.134	.209	-.991	.133	-1.012	.129	-.967	.129	-.978
-.311	-.453	.178	-1.025	.294	-.948	.214	-.989	.201	-.974	.209	-.949
-.023	-.533	.286	-.563	.404	-.807	.295	-.946	.294	-.928	.293	-.909
.133	-.427	.396	-.380	.497	-.573	.407	-.921	.397	-.895	.494	-.868
.272	-.359	.514	-.385	.599	-.518	.502	-.910	.495	-.843	.590	-.317
.416	-.205	.618	-.411	.700	-.544	.601	-.898	.594	-.511	.693	-.180
.565	-.128	.733	-.412	.864	-.213	.698	-.526	.693	-.358	.777	-.076
.713	-.143	.835	-.391	.926	-.157	.863	-.279	.784	-.203	.861	-.049
.854	-.227	.919	-.184			.923	-.200	.856	-.164	.918	-.046
.980	-.291	.987	-.080			.977	-.079	.926	-.133	.972	.007
1.074	-.360							.977	-.119		
					LOWER SURFACE						
-.660	.058	-.022	.404	.024	.392	.074	.104	.019	.291	.020	.317
-.616	.081	.038	.222	.075	.148	.130	.040	.066	.064	.076	.035
-.462	.024	.101	.126	.297	-.023	.298	-.032	.136	.003	.136	-.059
-.329	-.016	.185	.059	.400	-.052	.397	-.070	.214	-.023	.221	-.105
-.172	-.062	.398	-.020	.604	-.058	.501	-.067	.292	-.037	.295	-.131
-.030	-.100	.737	.136	.785	.188	.603	.032	.403	-.028	.396	-.142
.128	-.133			.967	.180	.784	.164	.489	-.088	.497	-.162
.418	-.025			1.000	-.087	.868	.257	.594	-.107	.597	-.101
.564	.012					.923	.314	.700	.039	.702	.038
.710	.113					.972	.187	.786	.151	.786	.132
.976	.267							.858	.232	.864	.182
1.072	.214							.919	.245	.912	.210
1.110	.137							.967	.085	.985	.077
CN=	.6322		.6927		.7232		.7968		.6649		.5484
CM=	-.0019		-.0630		-.0948		-.1384		-.0854		-.0454

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(e) M = 0.95. Continued.

$\alpha = 6.92^o$; $C_L = 0.789$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.201	-.021	-1.091	.023	-1.056	.025	-1.013	.022	-.948	.018	-.889
-.567	-.335	.035	-1.224	.068	-1.143	.079	-1.074	.075	-1.050	.077	-1.039
-.452	-.392	.105	-1.200	.209	-1.070	.133	-1.070	.129	-1.040	.129	-1.039
-.311	-.478	.178	-1.139	.294	-1.052	.214	-1.052	.201	-1.033	.209	-1.004
-.023	-.541	.286	-.847	.404	-.953	.295	-1.005	.294	-.988	.293	-.963
.133	-.432	.396	-.475	.497	-.737	.407	-.980	.397	-.956	.494	-.770
.272	-.366	.514	-.383	.599	-.629	.502	-.983	.495	-.882	.590	-.412
.416	-.279	.618	-.411	.700	-.545	.601	-.965	.594	-.559	.693	-.290
.565	-.154	.733	-.421	.864	-.285	.698	-.593	.693	-.288	.777	-.197
.713	-.173	.833	-.400	.926	-.212	.863	-.450	.784	-.241	.861	-.108
.854	-.254	.919	-.225			.923	-.386	.856	-.227	.918	-.105
.980	-.317	.987	-.105			.977	-.326	.926	-.223	.972	-.076
1.074	-.391							.977	-.223		
				LOWER SURFACE							
-.660	.062	-.022	.440	.024	.434	.074	.158	.019	.353	.020	.348
-.616	.100	.038	.281	.075	.207	.130	.083	.066	.134	.076	.066
-.462	.059	.101	.182	.297	.019	.298	.004	.136	.043	.136	-.020
-.329	.021	.185	.112	.400	-.017	.397	-.047	.214	.008	.221	-.080
-.172	-.030	.398	.020	.604	-.041	.501	-.060	.292	-.006	.295	-.117
-.030	-.063	.737	.147	.785	.184	.603	.029	.403	-.019	.396	-.160
.128	-.063			.967	.158	.784	.149	.489	-.084	.497	-.185
.418	.034			1.000	-.151	.868	.256	.594	-.131	.597	-.129
.564	.043					.923	.305	.700	.025	.702	.003
.710	.135					.972	.156	.786	.135	.786	.097
.976	.277							.858	.212	.864	.153
1.072	.234							.919	.223	.912	.159
1.110	.140							.967	.057	.985	-.020
CN=	.7343		.6009		.8299		.9035		.7205		.5855
CM=	.0045		-.0680		-.1104		-.1699		-.0892		-.0484

$\alpha = 8.01^o$; $C_L = 0.876$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.257	-.021	-1.155	.023	-1.131	.025	-1.098	.022	-1.030	.018	-.979
-.567	-.437	.035	-1.265	.068	-1.206	.079	-1.138	.075	-1.125	.077	-1.072
-.452	-.430	.105	-1.235	.209	-1.137	.133	-1.136	.129	-1.102	.129	-1.028
-.311	-.504	.178	-1.214	.294	-1.116	.214	-1.111	.201	-1.099	.209	-1.016
-.023	-.560	.286	-.939	.404	-.977	.295	-1.078	.294	-1.061	.293	-.805
.133	-.451	.396	-.757	.497	-.789	.407	-1.054	.397	-1.023	.494	-.536
.272	-.382	.514	-.555	.599	-.659	.502	-1.061	.495	-.964	.590	-.469
.416	-.293	.618	-.449	.700	-.554	.601	-.817	.594	-.633	.693	-.409
.565	-.184	.733	-.428	.864	-.366	.698	-.610	.693	-.422	.777	-.376
.713	-.202	.835	-.430	.926	-.329	.863	-.499	.784	-.339	.861	-.314
.854	-.291	.919	-.267			.923	-.413	.856	-.299	.918	-.293
.980	-.346	.987	-.136			.977	-.297	.926	-.320	.972	-.273
1.074	-.435							.977	-.299		
				LOWER SURFACE							
-.660	.067	-.022	.460	.024	.485	.074	.226	.019	.390	.020	.395
-.616	.119	.038	.329	.075	.277	.130	.128	.066	.165	.076	.113
-.462	.059	.101	.235	.297	.070	.298	.021	.136	.078	.136	.008
-.329	.060	.185	.163	.400	.017	.397	-.031	.214	.032	.221	-.052
-.172	.013	.398	.057	.604	-.034	.501	-.058	.292	.010	.295	-.091
-.030	-.022	.737	.160	.785	.171	.603	.022	.403	-.006	.396	-.147
.128	-.037			.967	.117	.784	.134	.489	-.076	.497	-.180
.418	.045			1.000	-.249	.868	.244	.594	-.115	.597	-.146
.564	.077					.923	.305	.700	.030	.702	-.020
.710	.162					.972	.164	.786	.132	.786	.057
.976	.294							.858	.211	.864	.116
1.072	.235							.919	.223	.912	.114
1.110	.143							.967	.042	.985	-.150
CN=	.8439		.9235		.9060		.9489		.8149		.5967
CM=	.0149		-.0850		-.1205		-.1673		-.1117		-.0638

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Concluded.

$\alpha = 8.55^{\circ}$; $C_L = 0.917$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.277	-.021	-1.190	.023	-1.158	.025	-1.125	.022	-1.076	.018	-.978
-.567	-.444	.035	-1.315	.068	-1.244	.079	-1.172	.075	-1.158	.077	-.950
-.452	-.437	.105	-1.250	.209	-1.169	.133	-1.163	.129	-1.137	.129	-.889
-.311	-.511	.178	-1.232	.294	-1.122	.214	-1.156	.201	-1.130	.209	-.741
-.023	-.559	.286	-.967	.404	-.949	.295	-1.105	.294	-1.093	.293	-.688
.133	-.450	.396	-.846	.497	-.793	.407	-1.093	.397	-1.043	.494	-.558
.272	-.385	.514	-.723	.599	-.674	.502	-1.072	.495	-.996	.590	-.479
.416	-.307	.618	-.619	.700	-.551	.601	-.817	.594	-.668	.693	-.425
.565	-.213	.733	-.458	.864	-.390	.698	-.594	.693	-.519	.777	-.379
.713	-.233	.835	-.442	.926	-.353	.863	-.419	.784	-.479	.861	-.343
.854	-.305	.919	-.300			.923	-.320	.856	-.433	.918	-.327
.980	-.363	.987	-.145			.977	-.235	.926	-.438	.972	-.305
1.074	-.437							.977	-.418		
					LOWER SURFACE						
-.660	.061	-.022	.481	.024	.509	.074	.244	.019	.413	.020	.412
-.616	.125	.038	.357	.075	.296	.130	.158	.066	.191	.076	.156
-.462	.110	.101	.266	.297	.086	.298	.040	.136	.096	.136	.037
-.329	.079	.185	.186	.400	.036	.397	-.023	.214	.050	.221	-.033
-.172	.044	.398	.082	.604	-.027	.501	-.055	.292	.019	.295	-.075
-.030	.006	.737	.172	.785	.168	.603	.017	.403	.002	.396	-.137
.128	-.010			.967	.104	.784	.125	.489	-.067	.497	-.172
.418	.074			1.000	-.310	.868	.237	.594	-.102	.597	-.149
.564	.100					.923	.297	.700	.038	.702	-.037
.710	.178					.972	.159	.786	.142	.786	.042
.976	.304							.858	.215	.864	.106
1.072	.225							.919	.222	.912	.103
1.110	.149							.967	.041	.985	-.197
CN=	.9033		1.0106		.9272		.9516		.8891		.5566
CM=	.0153		-.1035		-.1219		-.1530		-.1388		-.0675

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97.

$\alpha = -4.95^{\circ}$; $C_L = -0.560$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	.048	-.021	.396	.023	.371	.025	.331	.022	.430	.018	.460
	-.567	.017	.035	.204	.068	.218	.079	.201	.075	.226	.077	.239
	-.452	-.093	.105	.125	.209	.078	.133	.150	.129	.169	.129	.157
	-.311	-.206	.178	.092	.294	-.003	.214	.087	.201	.109	.209	.093
	-.023	-.133	.286	.032	.404	.013	.295	.069	.294	.070	.293	.027
	.133	.021	.396	.011	.497	-.024	.407	.021	.397	.040	.494	-.120
	.272	.038	.514	.006	.599	-.062	.502	-.032	.495	-.017	.590	-.195
	.416	.066	.618	-.034	.700	-.098	.601	-.084	.594	-.073	.693	-.300
	.565	.093	.733	-.061	.864	-.194	.698	-.166	.693	-.164	.777	-.400
	.713	.082	.835	-.119	.926	-.209	.863	-.209	.784	-.276	.861	-.506
	.854	.038	.919	-.116			.923	-.182	.856	-.397	.918	-.653
	.980	-.025	.987	-.005			.977	-.112	.926	-.583	.972	-.292
	1.074	-.096							.977	-.555		
						LOWER SURFACE						
	-.660	-.046	-.022	-.619	.024	-.942	.074	-1.149	.019	-1.036	.020	-.437
	-.616	-.145	.038	-.891	.075	-.975	.130	-1.110	.066	-.793	.076	-.439
	-.462	-.223	.101	-.827	.297	-.965	.298	-1.038	.136	-.939	.136	-.420
	-.329	-.228	.185	-.619	.400	-.945	.397	-.806	.214	-.762	.221	-.370
	-.172	-.259	.398	-.562	.604	-.280	.501	-.437	.292	-.729	.295	-.365
	-.030	-.324	.737	-.191	.785	-.048	.603	-.303	.403	-.667	.396	-.344
	.128	-.416			.967	.041	.784	-.007	.489	-.625	.497	-.320
	.418	-.256			1.000	-.051	.868	.062	.594	-.568	.597	-.306
	.564	-.295					.923	-.069	.700	-.469	.702	-.295
	.710	-.255					.972	-.079	.786	-.385	.786	-.275
	.976	.044							.858	-.402	.864	-.263
	1.072	.090							.919	-.379	.912	-.259
	1.110	.071							.967	-.387	.985	-.267
CN=	-.3518		-.5331		-.5209		-.5372		-.5449		-.2028	
CM=	-.0227		.0007		-.0271		-.0321		.0116		-.0405	

$\alpha = -4.02^{\circ}$; $C_L = -0.480$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	.046	-.021	.349	.023	.306	.025	.276	.022	.395	.018	.448
	-.567	.003	.035	.146	.068	.154	.079	.151	.075	.206	.077	.207
	-.452	-.114	.105	.081	.209	.028	.133	.104	.129	.144	.129	.142
	-.311	-.219	.178	.032	.294	-.052	.214	.059	.201	.080	.209	.073
	-.023	-.297	.286	.002	.404	-.020	.295	.047	.294	.057	.293	.013
	.133	-.007	.396	-.027	.497	-.058	.407	-.006	.397	.023	.494	-.118
	.272	.018	.514	-.020	.599	-.083	.502	-.047	.495	-.027	.590	-.190
	.416	.041	.618	-.050	.700	-.143	.601	-.094	.594	-.084	.693	-.287
	.565	.080	.733	-.084	.864	-.176	.698	-.172	.693	-.166	.777	-.389
	.713	.070	.835	-.136	.926	-.171	.863	-.209	.784	-.238	.861	-.485
	.854	.014	.919	-.112			.923	-.142	.856	-.347	.918	-.634
	.980	-.036	.987	.003			.977	-.032	.926	-.469	.972	-.275
	1.074	-.100							.977	-.544		
						LOWER SURFACE						
	-.660	-.027	-.022	-.526	.024	-.841	.074	-1.087	.019	-1.020	.020	-.447
	-.616	-.116	.038	-.755	.075	-.890	.130	-1.040	.066	-1.037	.076	-.431
	-.462	-.203	.101	-.638	.297	-.888	.298	-1.017	.136	-1.024	.136	-.392
	-.329	-.222	.185	-.606	.400	-.894	.397	-.964	.214	-.873	.221	-.368
	-.172	-.248	.398	-.530	.604	-.267	.501	-.615	.292	-.748	.295	-.337
	-.030	-.317	.737	-.133	.785	-.009	.603	-.192	.403	-.653	.396	-.302
	.128	-.435			.967	.088	.784	.177	.489	-.585	.497	-.305
	.418	-.250			1.000	.005	.868	.172	.594	-.537	.597	-.268
	.564	-.259					.923	.067	.700	-.458	.702	-.302
	.710	-.195					.972	.006	.786	-.337	.786	-.267
	.976	.061							.858	-.284	.864	-.255
	1.072	.110							.919	-.271	.912	-.260
	1.110	.087							.967	-.250	.985	-.266
CN=	-.2663		-.4397		-.4407		-.4830		-.5423		-.1876	
CM=	-.0167		-.0092		-.0337		-.0498		-.0022		-.0390	

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = -3.03^{\circ}$; $C_L = -0.380$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	.044	-.021	.264	.023	.230	.025	.213	.022	.342	.018	.422
-.567	-.010	.035	.064	.068	.071	.079	.086	.075	.145	.077	.165
-.452	-.135	.105	.028	.209	-.028	.133	.049	.129	.098	.129	.103
-.311	-.240	.178	-.021	.294	-.080	.214	.009	.201	.039	.209	.047
-.023	-.322	.286	-.056	.404	-.061	.295	.011	.294	.016	.293	-.010
.133	-.080	.396	-.069	.497	-.090	.407	-.032	.397	-.009	.494	-.129
.272	-.002	.514	-.055	.599	-.104	.502	-.074	.495	-.047	.590	-.192
.416	.025	.618	-.086	.700	-.181	.601	-.107	.594	-.091	.693	-.276
.565	.044	.733	-.110	.864	-.181	.698	-.179	.693	-.175	.777	-.372
.713	.045	.835	-.157	.926	-.103	.863	-.217	.784	-.267	.861	-.462
.854	-.009	.919	-.111			.923	-.136	.856	-.329	.918	-.613
.980	-.053	.987	.012			.977	.004	.926	-.380	.972	-.301
1.074	-.099							.977	-.204		
					LOWER SURFACE						
-.660	-.005	-.022	-.311	.024	-.759	.074	-1.022	.019	-.931	.020	-.440
-.616	-.080	.038	-.599	.075	-.800	.130	-.976	.066	-.985	.076	-.455
-.462	-.176	.101	-.564	.297	-.792	.298	-.963	.136	-.959	.136	-.382
-.329	-.202	.185	-.562	.400	-.758	.397	-.916	.214	-.801	.221	-.346
-.172	-.234	.398	-.480	.604	-.198	.501	-.577	.292	-.665	.295	-.333
-.030	-.307	.737	-.058	.785	.087	.603	-.057	.403	-.591	.396	-.291
.128	-.418			.967	.137	.784	.253	.489	-.558	.497	-.265
.418	-.257			1.000	.066	.868	.239	.594	-.503	.597	-.247
.564	-.264					.923	.183	.700	-.417	.702	-.249
.710	-.163					.972	.115	.786	-.278	.786	-.255
.976	.104							.858	-.176	.864	-.224
1.072	.148							.919	-.060	.912	-.222
1.110	.117							.967	.006	.985	-.221
CN=	-.1821		-.3185		-.3231		-.3831		-.4608		-.1565
CM=	-.0185		-.0233		-.0505		-.0722		-.0150		-.0457

$\alpha = -2.09^{\circ}$; $C_L = -0.276$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	.042	-.021	.176	.023	.151	.025	.145	.022	.291	.018	.383
-.567	-.028	.035	-.014	.068	-.005	.079	.019	.075	.091	.077	.125
-.452	-.158	.105	-.056	.209	-.082	.133	-.011	.129	.035	.129	.069
-.311	-.266	.178	-.078	.294	-.131	.214	-.044	.201	-.000	.209	.016
-.023	-.355	.286	-.104	.404	-.098	.295	-.033	.294	-.021	.293	-.036
.133	-.252	.396	-.106	.497	-.131	.407	-.059	.397	-.041	.494	-.153
.272	-.032	.514	-.090	.599	-.121	.502	-.099	.495	-.099	.590	-.218
.416	.005	.618	-.118	.700	-.178	.601	-.131	.594	-.107	.693	-.275
.565	.037	.733	-.136	.864	-.176	.698	-.200	.693	-.175	.777	-.274
.713	.031	.835	-.156	.926	-.084	.863	-.233	.784	-.285	.861	-.300
.854	-.025	.919	-.098			.923	-.138	.856	-.366	.918	-.441
.980	-.066	.987	.028			.977	.007	.926	-.136	.972	-.239
1.074	-.092							.977	.015		
					LOWER SURFACE						
-.660	.007	-.022	-.166	.024	-.654	.074	-.937	.019	-.844	.020	-.626
-.616	-.053	.038	-.442	.075	-.668	.130	-.917	.066	-.996	.076	-.600
-.462	-.146	.101	-.459	.297	-.727	.298	-.861	.136	-.919	.136	-.511
-.329	-.186	.185	-.488	.400	-.653	.397	-.829	.214	-.930	.221	-.449
-.172	-.216	.398	-.474	.604	-.151	.501	-.126	.292	-.695	.295	-.397
-.030	-.297	.737	-.004	.785	.133	.603	.033	.403	-.542	.396	-.310
.128	-.408			.967	.195	.784	.180	.489	-.492	.497	-.289
.418	-.312			1.000	.065	.868	.259	.594	-.385	.597	-.279
.564	-.250					.923	.226	.700	-.106	.702	-.222
.710	-.130					.972	.163	.786	.080	.786	-.223
.976	.127							.858	.230	.864	-.177
1.072	.162							.919	.223	.912	-.153
1.110	.138							.967	.192	.985	-.143
CN=	-.1071		-.2165		-.2211		-.2568		-.3282		-.2020
CM=	-.0080		-.0287		-.0601		-.0898		-.0690		-.0389

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = -1.20^{\circ}$; $C_L = -0.165$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	.034	-.021	-.000	.023	.041	.025	.066	.022	.220	.018	.311
	-.567	-.044	.035	-.098	.068	-.070	.079	-.046	.075	.035	.077	.059
	-.452	-.181	.105	-.123	.209	-.134	.133	-.070	.129	-.002	.129	.012
	-.311	-.289	.178	-.121	.294	-.187	.214	-.064	.201	-.046	.209	-.046
	-.023	-.383	.286	-.147	.404	-.132	.295	-.083	.294	-.054	.293	-.079
	.133	-.288	.396	-.152	.497	-.165	.407	-.085	.397	-.063	.494	-.183
	.272	-.207	.514	-.118	.599	-.165	.502	-.132	.495	-.125	.590	-.247
	.416	-.019	.618	-.145	.700	-.190	.601	-.170	.594	-.191	.693	-.342
	.565	.012	.733	-.146	.864	-.212	.698	-.209	.693	-.219	.777	-.408
	.713	.007	.835	-.163	.926	-.097	.863	-.286	.784	-.261	.861	-.067
	.854	-.040	.919	-.091			.923	-.158	.856	-.336	.918	-.044
	.980	-.077	.987	.038			.977	-.002	.926	-.205	.972	.036
	1.074	-.083							.977	-.017		
LOWER SURFACE												
	-.660	.019	-.022	-.050	.024	-.476	.074	-.861	.019	-.732	.020	-.786
	-.616	-.040	.038	-.346	.075	-.557	.130	-.830	.066	-.918	.076	-.718
	-.462	-.129	.101	-.419	.297	-.604	.298	-.757	.136	-.853	.136	-.680
	-.329	-.171	.185	-.421	.400	-.591	.397	-.226	.214	-.826	.221	-.619
	-.172	-.203	.398	-.447	.604	-.132	.501	-.151	.292	-.714	.295	-.526
	-.030	-.283	.737	.026	.785	.165	.603	.006	.403	-.302	.396	-.396
	.128	-.383			.967	.238	.784	.164	.489	-.210	.497	-.315
	.418	-.317			1.000	.049	.868	.225	.594	-.148	.597	-.219
	.564	-.265					.923	.249	.700	-.016	.702	-.140
	.710	-.097					.972	.171	.786	.078	.786	-.059
	.976	.152							.858	.175	.864	.002
	1.072	.186							.919	.255	.912	.060
	1.110	.155							.967	.137	.985	.057
CN=	-.0210		-.1291		-.1124		-.1371		-.1939		-.2101	
CM=	-.0048		-.0307		-.0708		-.0994		-.0909		-.0503	

(f) M = 0.97. Continued.

$\alpha = 0^{\circ}$; $C_L = -0.009$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	.011	-.021	-.136	.023	-.129	.025	-.113	.022	.111	.018	.210
	-.567	-.069	.035	-.374	.068	-.259	.079	-.164	.075	-.070	.077	-.048
	-.452	-.204	.105	-.319	.209	-.252	.133	-.182	.129	-.087	.129	-.089
	-.311	-.314	.178	-.344	.294	-.214	.214	-.160	.201	-.121	.209	-.113
	-.023	-.416	.286	-.167	.404	-.160	.295	-.141	.294	-.110	.293	-.146
	.133	-.326	.396	-.190	.497	-.201	.407	-.127	.397	-.117	.494	-.233
	.272	-.272	.514	-.164	.599	-.207	.502	-.182	.495	-.171	.590	-.280
	.416	-.116	.618	-.182	.700	-.237	.601	-.186	.594	-.233	.693	-.365
	.565	-.005	.733	-.171	.864	-.208	.698	-.272	.693	-.310	.777	-.401
	.713	-.007	.835	-.144	.926	-.119	.863	-.372	.784	-.361	.861	-.041
	.854	-.061	.919	-.080			.923	-.145	.856	-.296	.918	-.025
	.980	-.090	.987	.036			.977	.001	.926	-.205	.972	.064
	1.074	-.084							.977	-.024		
LOWER SURFACE												
	-.660	.035	-.022	.004	.024	-.270	.074	-.724	.019	-.557	.020	-.555
	-.616	-.023	.038	-.269	.075	-.446	.130	-.593	.066	-.777	.076	-.746
	-.462	-.106	.101	-.342	.297	-.420	.298	-.258	.136	-.694	.136	-.748
	-.329	-.147	.185	-.338	.400	-.169	.397	-.184	.214	-.590	.221	-.697
	-.172	-.187	.398	-.406	.604	-.183	.501	-.241	.292	-.281	.295	-.370
	-.030	-.267	.737	.096	.785	.173	.603	-.007	.403	-.280	.396	-.339
	.128	-.365			.967	.243	.784	.138	.489	-.185	.497	-.125
	.418	-.294			1.000	.040	.868	.200	.594	-.217	.597	.004
	.564	-.253					.923	.206	.700	.017	.702	.113
	.710	-.032					.972	.153	.786	.125	.786	.181
	.976	.189							.858	.181	.864	.213
	1.072	.214							.919	.215	.912	.221
	1.110	.172							.967	.143	.985	.113
CN=	.0842		.0160		.0589		.0195		-.0477		-.0406	
CM=	-.0073		-.0378		-.0762		-.0969		-.0955		-.0987	

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = 0.96^{\circ}$; $C_L = 0.110$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.009	-.021	-.340	.023	-.376	.025	-.243	.022	-.096	.018	.089
-.567	-.102	.035	-.588	.068	-.490	.079	-.347	.075	-.231	.077	-.155
-.452	-.220	.105	-.439	.209	-.257	.133	-.324	.129	-.217	.129	-.160
-.311	-.329	.178	-.401	.294	-.235	.214	-.252	.201	-.224	.209	-.176
-.023	-.443	.286	-.407	.404	-.218	.295	-.245	.294	-.203	.293	-.193
.133	-.359	.396	-.207	.497	-.248	.407	-.235	.397	-.127	.494	-.279
.272	-.305	.514	-.157	.599	-.282	.502	-.235	.495	-.172	.590	-.338
.416	-.242	.618	-.187	.700	-.321	.601	-.172	.594	-.253	.693	-.414
.565	-.065	.733	-.182	.864	-.185	.698	-.239	.693	-.340	.777	-.373
.713	-.020	.835	-.152	.926	-.088	.863	-.473	.784	-.417	.861	-.031
.854	-.071	.919	-.099			.923	-.166	.856	-.492	.918	-.032
.980	-.101	.987	.017			.977	-.012	.926	-.178	.972	.062
1.074	-.066							.977	-.046		
					LOWER SURFACE						
-.660	.043	-.022	.072	.024	-.099	.074	-.438	.019	-.358	.020	-.330
-.616	.005	.038	-.205	.075	-.328	.130	-.331	.066	-.556	.076	-.503
-.462	-.082	.101	-.267	.297	-.225	.298	-.241	.136	-.449	.136	-.470
-.329	-.119	.185	-.282	.400	-.239	.397	-.283	.214	-.374	.221	-.428
-.172	-.173	.398	-.254	.604	-.152	.501	-.241	.292	-.139	.295	-.342
-.030	-.257	.737	.110	.785	.183	.603	.007	.403	-.171	.396	-.365
.128	-.330			.967	.236	.784	.152	.489	-.153	.497	-.146
.418	-.270			1.000	.030	.868	.200	.594	-.241	.597	.007
.564	-.180					.923	.218	.700	.031	.702	.120
.710	-.002					.972	.170	.786	.138	.786	.173
.976	.214							.858	.206	.864	.218
1.072	.219							.919	.238	.912	.225
1.110	.178							.967	.146	.985	.111
CN=	.1907		.1580		.1706		.1429		.1070		.0673
CM=	-.0077		-.0409		-.0737		-.0943		-.1015		-.0919

(f) M = 0.97. Continued.

$\alpha = 1.93^{\circ}$; $C_L = 0.232$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.022	-.021	-.492	.023	-.534	.025	-.492	.022	-.272	.018	-.096
-.567	-.125	.035	-.707	.068	-.687	.079	-.537	.075	-.379	.077	-.303
-.452	-.234	.105	-.496	.209	-.514	.133	-.395	.129	-.352	.129	-.289
-.311	-.341	.178	-.445	.294	-.537	.214	-.208	.201	-.328	.209	-.247
-.023	-.452	.286	-.449	.404	-.178	.295	-.247	.294	-.275	.293	-.249
.133	-.368	.396	-.381	.497	-.223	.407	-.297	.397	-.200	.494	-.315
.272	-.318	.514	-.216	.599	-.308	.502	-.320	.495	-.221	.590	-.361
.416	-.267	.618	-.196	.700	-.319	.601	-.273	.594	-.287	.693	-.423
.565	-.098	.733	-.196	.864	-.204	.698	-.264	.693	-.373	.777	-.295
.713	-.054	.835	-.178	.926	-.105	.863	-.311	.784	-.464	.861	-.028
.854	-.090	.919	-.061			.923	-.129	.856	-.456	.918	-.021
.980	-.124	.987	.033			.977	-.017	.926	-.136	.972	.056
1.074	-.121							.977	-.059		
					LOWER SURFACE						
-.660	.052	-.022	.133	.024	.047	.074	-.193	.019	-.150	.020	-.080
-.616	.021	.038	-.106	.075	-.167	.130	-.175	.066	-.281	.076	-.325
-.462	-.058	.101	-.187	.297	-.189	.298	-.220	.136	-.220	.136	-.359
-.329	-.097	.185	-.219	.400	-.148	.397	-.129	.214	-.206	.221	-.342
-.172	-.159	.398	-.180	.604	-.060	.501	-.167	.292	-.209	.295	-.344
-.030	-.238	.737	.119	.785	.193	.603	.012	.403	-.195	.396	-.248
.128	-.304			.967	.237	.784	.157	.489	-.123	.497	-.182
.418	-.225			1.000	.033	.868	.224	.594	-.219	.597	-.020
.564	-.090					.923	.255	.700	.039	.702	.119
.710	.027					.972	.186	.786	.162	.786	.201
.976	.220							.858	.229	.864	.244
1.072	.222							.919	.257	.912	.249
1.110	.177							.967	.147	.985	.102
CN=	.2740		.2651		.3229		.2498		.2221		.1552
CM=	-.0140		-.0424		-.0757		-.0863		-.0976		-.0832

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = 2.47^{\circ}$; $C_L = 0.303$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.042	-.021	-.569	.023	-.609	.025	-.562	.022	-.433	.018	-.323
-.567	-.151	.035	-.741	.068	-.756	.079	-.660	.075	-.638	.077	-.530
-.452	-.259	.105	-.617	.209	-.654	.133	-.662	.129	-.576	.129	-.429
-.311	-.352	.178	-.492	.294	-.606	.214	-.574	.201	-.494	.209	-.358
-.023	-.467	.286	-.472	.404	-.160	.295	-.414	.294	-.399	.293	-.193
.133	-.385	.396	-.422	.497	-.248	.407	-.261	.397	-.194	.494	-.295
.272	-.332	.514	-.255	.599	-.318	.502	-.295	.495	-.218	.590	-.335
.416	-.287	.618	-.256	.700	-.354	.601	-.342	.594	-.234	.693	-.420
.565	-.121	.733	-.233	.864	-.191	.698	-.362	.693	-.279	.777	-.448
.713	-.075	.835	-.220	.926	-.090	.863	-.268	.784	-.372	.861	-.042
.854	-.113	.919	-.090			.923	-.155	.856	-.479	.918	-.009
.980	-.155	.987	.011			.977	-.008	.926	-.183	.972	.042
1.074	-.149							.977	-.056		
					LOWER SURFACE						
-.660	.059	-.022	.168	.024	.116	.074	-.117	.019	.006	.020	.093
-.616	.030	.038	-.044	.075	-.113	.130	-.149	.066	-.183	.076	-.184
-.462	-.046	.101	-.100	.297	-.170	.298	-.206	.136	-.161	.136	-.262
-.329	-.082	.185	-.125	.400	-.126	.397	-.121	.214	-.169	.221	-.235
-.172	-.150	.398	-.137	.604	-.050	.501	-.128	.292	-.168	.295	-.231
-.030	-.228	.737	.122	.785	.194	.603	.026	.403	-.064	.396	-.226
.128	-.290			.967	.242	.784	.173	.489	-.109	.497	-.157
.418	-.177			1.000	.023	.868	.245	.594	-.163	.597	-.038
.564	-.079					.923	.289	.700	.073	.702	.101
.710	.041					.972	.222	.786	.182	.786	.187
.976	.232							.858	.249	.864	.233
1.072	.230							.919	.267	.912	.242
1.110	.170							.967	.146	.985	.073
CN=	.3405		.3571		.3785		.3601		.3158		.2365
CM=	-.0165		-.0493		-.0740		-.0913		-.0891		-.0756

$\alpha = 2.88^{\circ}$; $C_L = 0.358$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.040	-.021	-.632	.023	-.644	.025	-.602	.022	-.485	.018	-.401
-.567	-.182	.035	-.763	.068	-.793	.079	-.711	.075	-.667	.077	-.649
-.452	-.269	.105	-.690	.209	-.735	.133	-.730	.129	-.664	.129	-.670
-.311	-.357	.178	-.508	.294	-.666	.214	-.662	.201	-.621	.209	-.585
-.023	-.474	.286	-.479	.404	-.416	.295	-.611	.294	-.583	.293	-.563
.133	-.382	.396	-.449	.497	-.200	.407	-.366	.397	-.515	.494	-.161
.272	-.336	.514	-.287	.599	-.297	.502	-.285	.495	-.261	.590	-.202
.416	-.288	.618	-.273	.700	-.356	.601	-.328	.594	-.323	.693	-.277
.565	-.143	.733	-.253	.864	-.213	.698	-.427	.693	-.325	.777	-.233
.713	-.082	.835	-.211	.926	-.100	.863	-.217	.784	-.264	.861	-.118
.854	-.124	.919	-.081			.923	-.115	.856	-.263	.918	-.105
.980	-.162	.987	.011			.977	.004	.926	-.225	.972	.029
1.074	-.178							.977	-.009		
					LOWER SURFACE						
-.660	.054	-.022	.204	.024	.154	.074	-.088	.019	.047	.020	.116
-.616	.031	.038	-.004	.075	-.075	.130	-.130	.066	-.172	.076	-.143
-.462	-.045	.101	-.068	.297	-.158	.298	-.196	.136	-.157	.136	-.224
-.329	-.083	.185	-.124	.400	-.102	.397	-.128	.214	-.164	.221	-.195
-.172	-.140	.398	-.136	.604	-.057	.501	-.127	.292	-.146	.295	-.189
-.030	-.221	.737	.116	.785	.189	.603	.027	.403	-.065	.396	-.185
.128	-.269			.967	.228	.784	.173	.489	-.106	.497	-.125
.418	-.130			1.000	.007	.868	.252	.594	-.120	.597	-.012
.564	-.074					.923	.295	.700	.084	.702	.120
.710	.048					.972	.223	.786	.193	.786	.191
.976	.230							.858	.263	.864	.241
1.072	.212							.919	.283	.912	.248
1.110	.162							.967	.179	.985	.099
CN=	.3751		.3824		.4302		.4109		.3951		.3082
CM=	-.0177		-.0471		-.0746		-.0896		-.0893		-.0597

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = 3.47^{\circ}$; $C_L = 0.432$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.057	-.021	-.695	.023	-.706	.025	-.662	.022	-.548	.018	-.479
-.567	-.193	.035	-.795	.068	-.858	.079	-.753	.075	-.735	.077	-.704
-.452	-.277	.105	-.775	.209	-.802	.133	-.763	.129	-.714	.129	-.728
-.311	-.378	.178	-.546	.294	-.735	.214	-.762	.201	-.732	.209	-.693
-.023	-.482	.286	-.501	.404	-.699	.295	-.704	.294	-.670	.293	-.654
.133	-.395	.396	-.488	.497	-.184	.407	-.660	.397	-.665	.494	-.625
.272	-.343	.514	-.291	.599	-.280	.502	-.297	.495	-.610	.590	-.117
.416	-.301	.618	-.280	.700	-.363	.601	-.339	.594	-.340	.693	-.001
.565	-.154	.733	-.257	.864	-.236	.698	-.432	.693	-.412	.777	-.072
.713	-.091	.835	-.241	.926	-.132	.863	-.203	.784	-.233	.861	-.134
.854	-.140	.919	-.108			.923	-.095	.856	-.214	.918	-.128
.980	-.182	.987	-.015			.977	-.017	.926	-.098	.972	.041
1.074	-.214							.977	.023		
				LOWER SURFACE							
-.660	.063	-.022	.287	.024	.201	.074	-.055	.019	.098	.020	.155
-.616	.045	.038	.056	.075	-.027	.130	-.101	.066	-.128	.076	-.123
-.462	-.022	.101	-.026	.297	-.124	.298	-.132	.136	-.131	.136	-.184
-.329	-.068	.185	-.079	.400	-.101	.397	-.123	.214	-.138	.221	-.176
-.172	-.131	.398	-.115	.604	-.059	.501	-.107	.292	-.120	.295	-.181
-.030	-.183	.737	.122	.785	.191	.603	.025	.403	-.057	.396	-.164
.128	-.263			.967	.219	.784	.168	.489	-.106	.497	-.138
.418	-.103			1.000	-.011	.868	.251	.594	-.114	.597	-.029
.564	-.055					.923	.298	.700	.088	.702	.120
.710	.061					.972	.210	.786	.199	.786	.207
.976	.241							.858	.272	.864	.249
1.072	.217							.919	.294	.912	.255
1.110	.160							.967	.191	.985	.143
CN=	.4300		.4397		.4990		.4833		.4822		.3753
CM=	-.0156		-.0493		-.0786		-.0923		-.0934		-.0559

$\alpha = 3.89^{\circ}$; $C_L = 0.487$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.081	-.021	-.773	.023	-.751	.025	-.711	.022	-.577	.018	-.527
-.567	-.222	.035	-.884	.068	-.880	.079	-.805	.075	-.770	.077	-.741
-.452	-.293	.105	-.867	.209	-.857	.133	-.807	.129	-.757	.129	-.749
-.311	-.394	.178	-.581	.294	-.789	.214	-.797	.201	-.761	.209	-.735
-.023	-.500	.286	-.516	.404	-.750	.295	-.752	.294	-.728	.293	-.699
.133	-.405	.396	-.507	.497	-.384	.407	-.730	.397	-.706	.494	-.665
.272	-.362	.514	-.339	.599	-.270	.502	-.471	.495	-.670	.590	-.594
.416	-.315	.618	-.311	.700	-.372	.601	-.369	.594	-.661	.693	-.085
.565	-.165	.733	-.289	.864	-.265	.698	-.419	.693	-.534	.777	-.022
.713	-.114	.835	-.253	.926	-.145	.863	-.211	.784	-.241	.861	-.018
.854	-.160	.919	-.116			.923	-.097	.856	-.166	.918	-.020
.980	-.205	.987	-.029			.977	-.040	.926	-.060	.972	.055
1.074	-.234							.977	.014		
				LOWER SURFACE							
-.660	.062	-.022	.324	.024	.246	.074	-.033	.019	.119	.020	.181
-.616	.056	.038	.109	.075	.005	.130	-.088	.066	-.095	.076	-.103
-.462	-.007	.101	-.006	.297	-.106	.298	-.099	.136	-.105	.136	-.170
-.329	-.057	.185	-.048	.400	-.101	.397	-.116	.214	-.116	.221	-.172
-.172	-.123	.398	-.098	.604	-.061	.501	-.102	.292	-.110	.295	-.178
-.030	-.167	.737	.124	.785	.190	.603	.021	.403	-.055	.396	-.163
.128	-.257			.967	.214	.784	.160	.489	-.102	.497	-.154
.418	-.092			1.000	-.022	.868	.247	.594	-.122	.597	-.050
.564	-.045					.923	.296	.700	.081	.702	.097
.710	.068					.972	.201	.786	.196	.786	.182
.976	.241							.858	.262	.864	.226
1.072	.216							.919	.287	.912	.235
1.110	.154							.967	.180	.985	.144
CN=	.4773		.4957		.5523		.5336		.5495		.4263
CM=	-.0128		-.0517		-.0846		-.0966		-.1051		-.0588

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = 4.91^{\circ}$; $C_L = 0.499$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	-.121	-.021	-.858	.023	-.853	.025	-.801	.022	-.695	.018	-.639
	-.567	-.259	.035	-1.021	.068	-.965	.079	-.870	.075	-.860	.077	-.830
	-.452	-.317	.105	-.965	.209	-.923	.133	-.884	.129	-.827	.129	-.842
	-.311	-.417	.178	-.887	.294	-.904	.214	-.874	.201	-.842	.209	-.810
	-.023	-.507	.286	-.536	.404	-.885	.295	-.827	.294	-.806	.293	-.776
	.133	-.414	.396	-.550	.497	-.607	.407	-.799	.397	-.782	.494	-.754
	.272	-.372	.514	-.384	.599	-.319	.502	-.798	.495	-.779	.590	-.694
	.416	-.313	.618	-.346	.700	-.408	.601	-.800	.594	-.764	.693	-.205
	.565	-.174	.733	-.312	.864	-.275	.698	-.520	.693	-.360	.777	-.173
	.713	-.134	.835	-.268	.926	-.141	.863	-.212	.784	-.265	.861	-.095
	.854	-.189	.919	-.133			.923	-.175	.856	-.229	.918	-.051
	.980	-.232	.987	-.056			.977	-.112	.926	-.205	.972	-.016
	1.074	-.271							.977	-.200		
						LOWER SURFACE						
	-.660	.073	-.022	.368	.024	.329	.074	.040	.019	.200	.020	.253
	-.616	.069	.038	.167	.075	.073	.130	-.019	.066	-.015	.076	-.044
	-.462	.015	.101	.074	.297	-.066	.298	-.067	.136	-.063	.136	-.132
	-.329	-.039	.185	.010	.400	-.072	.397	-.099	.214	-.088	.221	-.144
	-.172	-.090	.398	-.057	.604	-.056	.501	-.087	.292	-.083	.295	-.171
	-.030	-.135	.737	.138	.785	.195	.603	.027	.403	-.046	.396	-.192
	.128	-.214			.967	.202	.784	.149	.489	-.101	.497	-.230
	.418	-.055			1.000	-.060	.868	.243	.594	-.194	.597	-.123
	.564	-.008					.923	.296	.700	.041	.702	.021
	.710	.096					.972	.180	.786	.147	.786	.106
	.976	.259							.858	.221	.864	.173
	1.072	.230							.919	.234	.912	.175
	1.110	.156							.967	.087	.985	.038
CN=	.5609		.6067		.6541		.6840		.6070		.4869	
CM=	-.0108		-.0551		-.0925		-.1273		-.1040		-.0602	

$\alpha = 5.91^{\circ}$; $C_L = 0.698$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	-.139	-.021	-.974	.023	-.944	.025	-.881	.022	-.800	.018	-.716
	-.567	-.299	.035	-1.090	.068	-1.031	.079	-.956	.075	-.923	.077	-.908
	-.452	-.352	.105	-1.092	.209	-.983	.133	-.959	.129	-.912	.129	-.905
	-.311	-.432	.178	-1.052	.294	-.975	.214	-.944	.201	-.920	.209	-.880
	-.023	-.515	.286	-.618	.404	-.931	.295	-.897	.294	-.887	.293	-.845
	.133	-.431	.396	-.567	.497	-.896	.407	-.882	.397	-.861	.494	-.817
	.272	-.377	.514	-.419	.599	-.620	.502	-.879	.495	-.840	.590	-.795
	.416	-.318	.618	-.376	.700	-.452	.601	-.854	.594	-.634	.693	-.326
	.565	-.181	.733	-.341	.864	-.340	.698	-.875	.693	-.399	.777	-.262
	.713	-.157	.835	-.283	.926	-.190	.863	-.438	.784	-.333	.861	-.215
	.854	-.211	.919	-.144			.923	-.412	.856	-.263	.918	-.191
	.980	-.260	.987	-.076			.977	-.384	.926	-.233	.972	-.156
	1.074	-.319							.977	-.216		
						LOWER SURFACE						
	-.660	.074	-.022	.402	.024	.384	.074	.098	.019	.276	.020	.286
	-.616	.089	.038	.229	.075	.157	.130	.034	.066	.042	.076	-.017
	-.462	.042	.101	.133	.297	-.019	.298	-.029	.136	-.016	.136	-.102
	-.329	-.010	.185	.064	.400	-.048	.397	-.077	.214	-.054	.221	-.122
	-.172	-.057	.398	-.016	.604	-.044	.501	-.085	.292	-.056	.295	-.159
	-.030	-.102	.737	.145	.785	.196	.603	.027	.403	-.040	.396	-.201
	.128	-.149			.967	.182	.784	.141	.489	-.100	.497	-.241
	.418	-.021			1.000	-.075	.868	.231	.594	-.223	.597	-.168
	.564	.021					.923	.288	.700	.026	.702	-.027
	.710	.116					.972	.131	.786	.127	.786	.063
	.976	.275							.858	.203	.864	.123
	1.072	.237							.919	.220	.912	.131
	1.110	.155							.967	.047	.985	-.038
CN=	.6492		.7002		.7854		.8453		.6560		.5604	
CM=	-.0057		-.0576		-.1170		-.1813		-.1006		-.0754	

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{o}$

(f) M = 0.97. Continued.

$\alpha = 6.91^{o}$; $C_L = 0.792$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.173	-.021	-1.046	.023	-.990	.025	-.960	.022	-.891	.018	-.798
	-.567	-.336	.035	-1.150	.068	-1.086	.079	-1.020	.075	-.991	.077	-.952
	-.452	-.375	.105	-1.142	.209	-1.041	.133	-1.011	.129	-.980	.129	-.963
	-.311	-.454	.178	-1.135	.294	-1.030	.214	-1.004	.201	-.967	.209	-.938
	-.023	-.516	.286	-.852	.404	-.997	.295	-.958	.294	-.936	.293	-.904
	.133	-.433	.396	-.607	.497	-.977	.407	-.947	.397	-.916	.494	-.881
	.272	-.378	.514	-.467	.599	-.831	.502	-.931	.495	-.887	.590	-.848
	.416	-.322	.618	-.406	.700	-.620	.601	-.925	.594	-.721	.693	-.339
	.565	-.214	.733	-.363	.864	-.314	.698	-.632	.693	-.479	.777	-.312
	.713	-.182	.835	-.305	.926	-.192	.863	-.451	.784	-.316	.861	-.309
	.854	-.235	.919	-.169			.923	-.450	.856	-.261	.918	-.309
	.980	-.281	.987	-.090			.977	-.419	.926	-.239	.972	-.301
	1.074	-.347							.977	-.239		
LOWER SURFACE												
	-.660	.071	-.022	.436	.024	.440	.074	.157	.019	.332	.020	.324
	-.616	.105	.038	.284	.075	.218	.130	.082	.066	.093	.076	.036
	-.462	.065	.101	.181	.297	.024	.298	-.006	.136	.019	.136	-.059
	-.329	.021	.185	.107	.400	-.014	.397	-.052	.214	-.019	.221	-.098
	-.172	-.032	.398	.022	.604	-.035	.501	-.073	.292	-.038	.295	-.142
	-.030	-.064	.737	.159	.785	.194	.603	.019	.403	-.033	.396	-.184
	.128	-.079			.967	.174	.784	.104	.489	-.099	.497	-.250
	.418	.022			1.000	-.172	.868	.222	.594	-.219	.597	-.186
	.564	.052					.923	.268	.700	.009	.702	-.044
	.710	.139					.972	.114	.786	.117	.786	.046
	.976	.289							.858	.197	.864	.103
	1.072	.239							.919	.206	.912	.110
	1.110	.159							.967	.034	.985	-.087
CN=	.7387		.7976		.8835		.8694		.7126		.6274	
CM=	-.0011		-.0654		-.1332		-.1680		-.1034		-.0880	

$\alpha = 7.98^{o}$; $C_L = 0.878$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.248	-.021	-1.096	.023	-1.053	.025	-1.028	.022	-.956	.018	-.882
	-.567	-.394	.035	-1.217	.068	-1.146	.079	-1.088	.075	-1.058	.077	-1.021
	-.452	-.407	.105	-1.211	.209	-1.091	.133	-1.075	.129	-1.038	.129	-1.029
	-.311	-.477	.178	-1.162	.294	-1.072	.214	-1.055	.201	-1.034	.209	-.996
	-.023	-.539	.286	-.985	.404	-1.045	.295	-1.023	.294	-1.001	.293	-.971
	.133	-.443	.396	-.812	.497	-.896	.407	-1.014	.397	-.972	.494	-.936
	.272	-.367	.514	-.633	.599	-.903	.502	-1.007	.495	-.937	.590	-.740
	.416	-.330	.618	-.455	.700	-.594	.601	-.967	.594	-.829	.693	-.394
	.565	-.243	.733	-.395	.864	-.351	.698	-.626	.693	-.490	.777	-.381
	.713	-.207	.835	-.342	.926	-.278	.863	-.543	.784	-.344	.861	-.379
	.854	-.270	.919	-.231			.923	-.493	.856	-.305	.918	-.367
	.980	-.307	.987	-.123			.977	-.444	.926	-.322	.972	-.361
	1.074	-.370							.977	-.327		
LOWER SURFACE												
	-.660	.072	-.022	.454	.024	.482	.074	.213	.019	.374	.020	.366
	-.616	.123	.038	.337	.075	.270	.130	.125	.066	.136	.076	.094
	-.462	.092	.101	.236	.297	.068	.298	.021	.136	.058	.136	-.013
	-.329	.057	.185	.161	.400	.019	.397	-.043	.214	.010	.221	-.075
	-.172	.023	.398	.059	.604	-.027	.501	-.077	.292	-.013	.295	-.117
	-.030	-.019	.737	.166	.785	.174	.603	.011	.403	-.018	.396	-.163
	.128	-.047			.967	.113	.784	.108	.489	-.086	.497	-.227
	.418	.053			1.000	-.308	.868	.233	.594	-.202	.597	-.186
	.564	.065					.923	.287	.700	.015	.702	-.057
	.710	.169					.972	.136	.786	.116	.786	.034
	.976	.300							.858	.197	.864	.090
	1.072	.247							.919	.208	.912	.092
	1.110	.158							.967	.023	.985	-.151
CN=	.8369		.9198		.9278		.9399		.7882		.6832	
CM=	.0101		-.0830		-.1330		-.1797		-.1171		-.0920	

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = 9.08^{\circ}$; $C_L = 0.954$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	-.323	-.021	-1.146	.023	-1.126	.025	-1.098	.022	-1.023	.018	-.948
-.567	-.442	.035	-1.261	.068	-1.200	.079	-1.138	.075	-1.114	.077	-1.072
-.452	-.443	.105	-1.215	.209	-1.123	.133	-1.128	.129	-1.089	.129	-1.064
-.311	-.492	.178	-1.204	.294	-1.108	.214	-1.107	.201	-1.083	.209	-1.030
-.023	-.546	.286	-1.011	.404	-1.017	.295	-1.070	.294	-1.046	.293	-.986
.133	-.451	.396	-.897	.497	-.830	.407	-1.068	.397	-1.024	.494	-.748
.272	-.389	.514	-.828	.599	-.696	.502	-1.038	.495	-1.006	.590	-.540
.416	-.344	.618	-.750	.700	-.579	.601	-.868	.594	-.963	.693	-.478
.565	-.273	.733	-.515	.864	-.442	.698	-.667	.693	-.717	.777	-.434
.713	-.237	.835	-.439	.926	-.432	.863	-.496	.784	-.547	.861	-.409
.854	-.297	.919	-.313			.923	-.426	.856	-.491	.918	-.397
.980	-.329	.987	-.154			.977	-.337	.926	-.469	.972	-.382
1.074	-.397							.977	-.453		
LOWER SURFACE											
-.660	.065	-.022	.484	.024	.525	.074	.254	.019	.418	.020	.408
-.616	.140	.038	.381	.075	.332	.130	.165	.066	.186	.076	.137
-.462	.119	.101	.288	.297	.104	.298	.044	.136	.092	.136	.030
-.329	.095	.185	.213	.400	.051	.397	-.025	.214	.039	.221	-.054
-.172	.056	.398	.113	.604	-.019	.501	-.076	.292	.011	.295	-.088
-.030	.026	.737	.184	.785	.159	.603	.015	.403	-.008	.396	-.148
.128	.006			.967	.088	.784	.091	.489	-.074	.497	-.210
.418	.093			1.000	-.404	.868	.227	.594	-.190	.597	-.181
.564	.116					.923	.281	.700	.019	.702	-.066
.710	.195					.972	.132	.786	.124	.786	.013
.976	.321							.858	.200	.864	.066
1.072	.267							.919	.206	.912	.064
1.110	.172							.967	.013	.985	-.246
CN=	.9355		1.0667		.9548		.9593		.9136		.6707
CM=	.0161		-.1215		-.1345		-.1665		-.1591		-.0786

$\alpha = 10.04^{\circ}$; $C_L = 1.016$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	-.349	-.021	-1.187	.023	-1.174	.025	-1.141	.022	-1.081	.018	-1.014
-.567	-.492	.035	-1.263	.068	-1.226	.079	-1.175	.075	-1.155	.077	-1.017
-.452	-.481	.105	-1.251	.209	-1.166	.133	-1.166	.129	-1.137	.129	-1.050
-.311	-.516	.178	-1.176	.294	-1.082	.214	-1.144	.201	-1.129	.209	-.962
-.023	-.551	.286	-1.039	.404	-.910	.295	-1.104	.294	-1.090	.293	-.787
.133	-.447	.396	-.965	.497	-.797	.407	-1.022	.397	-1.057	.494	-.666
.272	-.404	.514	-.938	.599	-.690	.502	-1.015	.495	-1.020	.590	-.595
.416	-.356	.618	-.898	.700	-.608	.601	-.843	.594	-.864	.693	-.552
.565	-.290	.733	-.795	.864	-.487	.698	-.748	.693	-.704	.777	-.512
.713	-.261	.835	-.535	.926	-.504	.863	-.465	.784	-.623	.861	-.463
.854	-.310	.919	-.309			.923	-.370	.856	-.554	.918	-.444
.980	-.356	.987	-.179			.977	-.324	.926	-.486	.972	-.428
1.074	-.422							.977	-.497		
LOWER SURFACE											
-.660	.069	-.022	.502	.024	.559	.074	.306	.019	.448	.020	.434
-.616	.161	.038	.417	.075	.380	.130	.203	.066	.226	.076	.174
-.462	.154	.101	.331	.297	.155	.298	.070	.136	.129	.136	.054
-.329	.128	.185	.250	.400	.081	.397	.001	.214	.064	.221	-.024
-.172	.092	.398	.132	.604	-.007	.501	-.052	.292	.021	.295	-.079
-.030	.064	.737	.198	.785	.158	.603	.021	.403	.006	.396	-.136
.128	.046			.967	.065	.784	.066	.489	-.072	.497	-.198
.418	.123			1.000	-.498	.868	.215	.594	-.195	.597	-.178
.564	.150					.923	.274	.700	.013	.702	-.066
.710	.213					.972	.102	.786	.118	.786	.004
.976	.348							.858	.199	.864	.056
1.072	.276							.919	.202	.912	.052
1.110	.183							.967	.012	.985	-.301
CN=	1.0235		1.1732		.9841		.9794		.9461		.6582
CM=	.0247		-.1539		-.1390		-.1597		-.1612		-.0853

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Concluded.

$\alpha = 11.66^{\circ}$; $C_L = 1.106$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.458	-.021	-1.262	.023	-1.117	.025	-1.174	.022	-1.027	.018	-1.086
-.567	-.580	.035	-1.244	.068	-1.105	.079	-1.164	.075	-1.040	.077	-1.115
-.452	-.539	.105	-1.250	.209	-1.002	.133	-1.118	.129	-.981	.129	-1.073
-.311	-.560	.178	-1.191	.294	-.924	.214	-1.008	.201	-.954	.209	-.974
-.023	-.569	.286	-1.152	.404	-.827	.295	-.802	.294	-.889	.293	-.861
.133	-.472	.396	-1.145	.497	-.799	.407	-.435	.397	-.858	.494	-.776
.272	-.427	.514	-1.154	.599	-.741	.502	-.435	.495	-.876	.590	-.708
.416	-.402	.618	-1.155	.700	-.692	.601	-.467	.594	-.874	.693	-.661
.565	-.339	.733	-1.139	.864	-.646	.698	-.489	.693	-.826	.777	-.627
.713	-.292	.835	-.848	.926	-.607	.863	-.457	.784	-.678	.861	-.555
.854	-.346	.919	-.317			.923	-.453	.856	-.660	.918	-.534
.980	-.383	.987	-.130			.977	-.455	.926	-.695	.972	-.527
1.074	-.468							.977	-.657		
					LOWER SURFACE						
-.660	.062	-.022	.500	.024	.602	.074	.375	.019	.497	.020	.473
-.616	.185	.038	.474	.075	.439	.130	.262	.066	.297	.076	.234
-.462	.205	.101	.397	.297	.199	.298	.117	.136	.188	.136	.110
-.329	.138	.185	.319	.400	.141	.397	.038	.214	.118	.221	.024
-.172	.154	.398	.196	.604	.026	.501	-.008	.292	.071	.295	-.035
-.030	.127	.737	.227	.785	.172	.603	.043	.403	.033	.396	-.105
.128	.101			.967	.064	.784	.062	.489	-.050	.497	-.181
.418	.175			1.000	-.589	.868	.219	.594	-.171	.597	-.179
.564	.187					.923	.276	.700	.011	.702	-.086
.710	.267					.972	.089	.786	.117	.786	-.009
.976	.280							.858	.192	.864	.048
1.072	.302							.919	.204	.912	.038
1.110	.197							.967	-.002	.985	-.331
CN=	1.1841		1.3790		1.0084		.7901		.9294		.7604
CM=	.0402		-.2131		-.1691		-.1148		-.1832		-.1084

[redacted]

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(g) M = 0.98.

$\alpha = -1.06^{\circ}$; $C_L = -0.148$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.044	-.021	-.012	.023	-.012	.025	.058	.022	.245	.018	.327
-.567	-.038	.035	-.245	.068	-.155	.079	-.076	.075	.022	.077	.059
-.452	-.160	.105	-.255	.134	-.128	.133	-.085	.129	-.015	.129	.000
-.311	-.265	.178	-.322	.209	-.139	.214	-.081	.201	-.036	.209	-.037
-.023	-.367	.286	-.155	.294	-.146	.295	-.083	.294	-.046	.293	-.068
.133	-.277	.396	-.153	.404	-.131	.407	-.072	.397	-.068	.494	-.169
.272	-.238	.514	-.115	.497	-.167	.502	-.111	.495	-.124	.590	-.230
.416	-.138	.618	-.131	.599	-.184	.601	-.159	.594	-.182	.693	-.330
.565	.008	.733	-.142	.700	-.174	.698	-.225	.693	-.248	.777	-.418
.713	.013	.835	-.160	.864	-.172	.863	-.264	.784	-.255	.861	-.085
.854	-.034	.919	-.071	.926	-.074	.923	-.133	.856	-.355	.918	-.022
.980	-.070	.987	.055	.975	.026	.977	.012	.926	-.190	.972	.044
1.074	-.076							.977	-.010		
1.122	-.010										
				LOWER SURFACE							
-.660	.031	-.022	-.080	.024	-.403	.025	-.615	.019	-.687	.020	-.752
-.616	-.022	.038	-.340	.075	-.515	.130	-.815	.066	-.872	.076	-.682
-.572	-.075	.101	-.406	.297	-.575	.298	-.720	.136	-.820	.136	-.749
-.462	-.109	.185	-.431	.400	-.569	.397	-.417	.214	-.804	.221	-.602
-.329	-.126	.398	-.442	.604	-.125	.501	-.058	.292	-.633	.295	-.519
-.172	-.189	.737	.024	.785	.191	.603	.039	.403	-.323	.396	-.398
-.030	-.266			.967	.226	.703	.112	.489	-.222	.497	-.274
.128	-.384			1.000	.056	.784	.174	.594	-.135	.597	-.216
.418	-.310					.868	.231	.700	-.021	.702	-.068
.564	-.319					.923	.245	.786	.088	.786	-.058
.710	-.152					.972	.177	.858	.164	.864	.001
.976	.133							.919	.198	.912	.083
1.072	.172							.967	.122		
1.110	.150										
CN=	-.0257		-.0976		-.0967		-.1157		-.1829		-.2023
CM=	.0087		-.0232		-.0619		-.0954		-.0877		-.0544

$\alpha = -0.08^{\circ}$; $C_L = -0.235$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.017	-.021	-.125	.023	-.262	.025	-.058	.022	.162	.018	.243
-.567	-.064	.035	-.404	.068	-.387	.079	-.164	.075	-.053	.077	-.033
-.452	-.182	.105	-.332	.134	-.348	.133	-.177	.129	-.073	.129	-.074
-.311	-.291	.178	-.409	.209	-.184	.214	-.149	.201	-.096	.209	-.109
-.023	-.391	.286	-.348	.294	-.098	.295	-.154	.294	-.095	.293	-.126
.133	-.311	.396	-.213	.404	-.153	.407	-.069	.397	-.115	.494	-.239
.272	-.271	.514	-.125	.497	-.208	.502	-.128	.495	-.166	.590	-.274
.416	-.222	.618	-.134	.599	-.222	.601	-.168	.594	-.221	.693	-.368
.565	-.069	.733	-.139	.700	-.255	.698	-.247	.693	-.311	.777	-.438
.713	-.033	.835	-.167	.864	-.144	.863	-.414	.784	-.370	.861	-.098
.854	-.061	.919	-.072	.926	-.068	.923	-.141	.856	-.305	.918	.007
.980	-.080	.987	.052	.975	.002	.977	.004	.926	-.183	.972	.071
1.074	-.068							.977	-.021		
1.122	-.012										
				LOWER SURFACE							
-.660	.034	-.022	.012	.024	-.229	.025	-.504	.019	-.524	.020	-.563
-.616	-.014	.038	-.261	.075	-.440	.130	-.654	.066	-.751	.076	-.727
-.572	-.047	.101	-.358	.297	-.490	.298	-.099	.136	-.694	.136	-.747
-.462	-.059	.185	-.363	.400	-.516	.397	-.145	.214	-.377	.221	-.722
-.329	-.114	.398	-.425	.604	-.109	.501	-.215	.292	-.312	.295	-.564
-.172	-.176	.737	.054	.785	.191	.603	.003	.403	-.258	.396	-.316
-.030	-.262			.967	.249	.703	.088	.489	-.193	.497	-.139
.128	-.357			1.000	.054	.784	.161	.594	-.196	.597	-.001
.418	-.296					.868	.208	.700	.005	.702	.104
.564	-.290					.923	.228	.786	.102	.786	.165
.710	-.130					.972	.170	.858	.167	.864	.198
.976	.155							.919	.191	.912	.203
1.072	.184							.967	.128		
1.110	.155										
CN=	.0671		.0052		.0070		.0517		-.0439		-.0591
CM=	.0060		-.0241		-.0621		-.1002		-.0915		-.1012

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(g) M = 0.98. *Continued.*

$\alpha = 0.96^{\circ}$; $C_L = 0.114$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .48C X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.002	-.021	-.341	.023	-.385	.025	-.188	.022	-.005	.018	.105
-.567	-.106	.035	-.574	.068	-.530	.079	-.318	.075	-.210	.077	-.167
-.452	-.208	.105	-.424	.134	-.450	.133	-.181	.129	-.204	.129	-.175
-.311	-.302	.178	-.462	.209	-.469	.214	-.146	.201	-.211	.209	-.170
-.023	-.414	.286	-.411	.294	-.449	.295	-.154	.294	-.184	.293	-.180
.133	-.343	.396	-.347	.404	-.188	.407	-.183	.397	-.173	.494	-.267
.272	-.296	.514	-.238	.497	-.182	.502	-.213	.495	-.222	.590	-.309
.416	-.259	.618	-.245	.599	-.120	.601	-.249	.594	-.254	.693	-.382
.565	-.136	.733	-.204	.700	-.147	.698	-.312	.693	-.341	.777	-.450
.713	-.076	.835	-.077	.864	-.262	.863	-.294	.784	-.430	.861	-.045
.854	-.125	.919	-.016	.926	-.123	.923	-.097	.856	-.315	.918	.010
.980	-.136	.987	.064	.975	-.031	.977	.007	.926	-.185	.972	.077
1.074	-.077							.977	-.034		
1.122	-.003										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .48C X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.054	-.022	.069	.024	-.133	.025	-.239	.019	-.361	.020	-.303
-.616	.010	.038	-.211	.075	-.362	.130	-.191	.066	-.526	.076	-.544
-.572	-.031	.101	-.280	.297	-.436	.298	-.248	.136	-.435	.136	-.498
-.462	-.073	.185	-.275	.400	.042	.397	-.295	.214	-.342	.221	-.437
-.329	-.096	.398	-.382	.604	-.074	.501	-.223	.292	-.125	.295	-.312
-.172	-.161	.737	.111	.785	.194	.603	.025	.403	-.240	.396	-.387
-.030	-.248			.967	.236	.703	.100	.489	-.141	.497	-.192
.128	-.335			1.000	.030	.784	.152	.594	-.232	.597	.000
.418	-.268					.868	.207	.700	.043	.702	.114
.564	-.245					.923	.224	.786	.162	.786	.174
.710	-.064					.972	.173	.858	.224	.864	.204
.976	.186							.919	.251	.912	.210
1.072	.207							.967	.158		
1.110	.181										
CN= .1812		.1442		.1939		.1378		.1030		.0563	
CM= -.0082		-.0338		-.0724		-.0830		-.0998		-.0913	

$\alpha = 1.96^{\circ}$; $C_L = 0.245$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .48C X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.026	-.021	-.458	.023	-.497	.025	-.492	.022	-.321	.018	-.073
-.567	-.149	.035	-.711	.068	-.681	.079	-.602	.075	-.534	.077	-.289
-.452	-.237	.105	-.506	.134	-.586	.133	-.608	.129	-.285	.129	-.283
-.311	-.330	.178	-.539	.209	-.541	.214	-.526	.201	-.383	.209	-.252
-.023	-.440	.286	-.458	.294	-.502	.295	-.482	.294	-.107	.293	-.249
.133	-.375	.396	-.433	.404	-.461	.407	-.165	.397	-.128	.494	-.302
.272	-.324	.514	-.303	.497	-.249	.502	-.224	.495	-.224	.590	-.353
.416	-.294	.618	-.286	.599	-.305	.601	-.122	.594	-.291	.693	-.429
.565	-.180	.733	-.275	.700	-.298	.698	-.204	.693	-.367	.777	-.518
.713	-.119	.835	-.221	.864	-.174	.863	-.395	.784	-.440	.861	-.049
.854	-.160	.919	-.049	.926	-.078	.923	-.186	.856	-.527	.918	.013
.980	-.193	.987	.031	.975	-.003	.977	-.008	.926	-.174	.972	.043
1.074	-.176							.977	-.092		
1.122	-.067										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .48C X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.062	-.022	.137	.024	-.014	.025	-.067	.019	-.036	.020	-.016
-.616	.021	.038	-.139	.075	-.237	.130	-.150	.066	-.227	.076	-.299
-.572	-.004	.101	-.190	.297	-.202	.298	-.204	.136	-.180	.136	-.323
-.462	-.051	.185	-.246	.400	-.202	.397	-.128	.214	-.168	.221	-.316
-.329	-.081	.398	-.267	.604	-.072	.501	-.109	.292	-.184	.295	-.331
-.172	-.144	.737	.111	.785	.202	.603	.037	.403	-.051	.396	-.254
-.030	-.225			.967	.255	.703	.101	.489	-.105	.497	-.240
.128	-.298			1.000	.055	.784	.175	.594	-.213	.597	-.037
.418	-.247					.868	.246	.700	.046	.702	.110
.564	-.180					.923	.268	.786	.162	.786	.182
.710	-.002					.972	.205	.858	.235	.864	.225
.976	.211							.919	.252	.912	.237
1.072	.210							.967	.133		
1.110	.166										
CN= .3029		.2759		.3154		.3131		.2491		.1750	
CM= -.0207		-.0505		-.0758		-.0818		-.0992		-.0914	

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(g) M = 0.98. Continued.

$\alpha = 2.47^o$; $C_L = 0.313$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.028	-.021	-.573	.023	-.574	.025	-.546	.022	-.401	.018	-.292
-.567	-.160	.035	-.732	.068	-.739	.079	-.637	.075	-.624	.077	-.541
-.452	-.252	.105	-.654	.134	-.717	.133	-.668	.129	-.610	.129	-.392
-.311	-.343	.178	-.545	.209	-.650	.214	-.655	.201	-.614	.209	-.487
-.023	-.446	.286	-.476	.294	-.598	.295	-.593	.294	-.546	.293	-.139
.133	-.371	.396	-.467	.404	-.479	.407	-.398	.397	-.016	.494	-.265
.272	-.330	.514	-.329	.497	-.275	.502	-.215	.495	-.133	.590	-.333
.416	-.302	.618	-.300	.599	-.310	.601	-.222	.594	-.192	.693	-.392
.565	-.202	.733	-.284	.700	-.374	.698	-.204	.693	-.270	.777	-.485
.713	-.125	.835	-.231	.864	-.132	.863	-.301	.784	-.367	.861	-.063
.854	-.162	.919	-.064	.926	-.054	.923	-.131	.856	-.373	.918	.005
.980	-.195	.987	.016	.975	-.006	.977	.005	.926	-.214	.972	.054
1.074	-.196							.977	-.048		
1.122	-.082										
				LOWER SURFACE							
-.660	.062	-.022	.171	.024	.109	.025	.007	.019	.001	.020	.095
-.616	.035	.038	-.085	.075	-.118	.130	-.140	.066	-.177	.076	-.183
-.572	.001	.101	-.151	.297	-.176	.298	-.209	.136	-.171	.136	-.246
-.462	-.037	.185	-.211	.400	-.154	.397	-.107	.214	-.163	.221	-.234
-.329	-.073	.398	-.148	.604	-.049	.501	-.135	.292	-.170	.295	-.201
-.172	-.132	.737	.115	.785	.196	.603	.038	.403	-.046	.396	-.186
-.030	-.224			.967	.246	.703	.098	.489	-.103	.497	-.128
.128	-.287			1.000	.038	.784	.179	.594	-.152	.597	-.043
.418	-.221					.868	.253	.700	.076	.702	.101
.564	-.127					.923	.295	.786	.194	.786	.189
.710	.033					.972	.226	.858	.254	.864	.232
.976	.220							.919	.277	.912	.241
1.072	.218							.967	.158		
1.110	.161										

	.133	.307	.480	.653	.804	.933
CN=	.3492	.3573	.3944	.3707	.3081	.2508
CM=	-.0245	-.0554	-.0728	-.0789	-.0817	-.0811

$\alpha = 2.94^o$; $C_L = 0.378$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.047	-.021	-.643	.023	-.626	.025	-.597	.022	-.459	.018	-.384
-.567	-.182	.035	-.770	.068	-.787	.079	-.696	.075	-.679	.077	-.621
-.452	-.268	.105	-.701	.134	-.735	.133	-.708	.129	-.654	.129	-.655
-.311	-.367	.178	-.574	.209	-.701	.214	-.717	.201	-.673	.209	-.616
-.023	-.455	.286	-.481	.294	-.666	.295	-.636	.294	-.607	.293	-.586
.133	-.387	.396	-.487	.404	-.529	.407	-.624	.397	-.573	.494	-.301
.272	-.343	.514	-.344	.497	-.413	.502	-.352	.495	-.225	.590	-.103
.416	-.310	.618	-.322	.599	-.319	.601	-.295	.594	-.213	.693	-.051
.565	-.208	.733	-.299	.700	-.418	.698	-.233	.693	-.301	.777	-.140
.713	-.137	.835	-.254	.864	-.146	.863	-.271	.784	-.353	.861	-.189
.854	-.171	.919	-.090	.926	-.076	.923	-.130	.856	-.254	.918	-.121
.980	-.205	.987	-.004	.975	-.013	.977	-.002	.926	-.119	.972	.041
1.074	-.217							.977	.008		
1.122	-.116										
				LOWER SURFACE							
-.660	.069	-.022	.203	.024	.139	.025	.078	.019	.045	.020	.114
-.616	.049	.038	-.022	.075	-.093	.130	-.119	.066	-.134	.076	-.147
-.572	.017	.101	-.127	.297	-.157	.298	-.191	.136	-.153	.136	-.214
-.462	-.021	.185	-.157	.400	-.139	.397	-.089	.214	-.146	.221	-.202
-.329	-.060	.398	-.123	.604	-.044	.501	-.154	.292	-.155	.295	-.198
-.172	-.125	.737	.121	.785	.198	.603	.034	.403	-.048	.396	-.177
-.030	-.213			.967	.235	.703	.090	.489	-.093	.497	-.152
.128	-.268			1.000	.012	.784	.179	.594	-.148	.597	-.016
.418	-.189					.868	.254	.700	.088	.702	.133
.564	-.082					.923	.293	.786	.199	.786	.213
.710	.039					.972	.225	.858	.267	.864	.260
.976	.230							.919	.294	.912	.259
1.072	.219							.967	.191		
1.110	.156										

	.133	.307	.480	.653	.804	.933
CN=	.4043	.4108	.4500	.4422	.3944	.3077
CM=	-.0210	-.0593	-.0801	-.0846	-.0837	-.0588

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(g) M = 0.98. Continued.

$\alpha = 3.46^{\circ}$; $C_L = 0.444$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.054	-.021	-.686	.023	-.676	.025	-.643	.022	-.518	.018	-.452
-.567	-.205	.035	-.806	.068	-.824	.079	-.744	.075	-.725	.077	-.666
-.452	-.270	.105	-.770	.134	-.789	.133	-.742	.129	-.695	.129	-.701
-.311	-.362	.178	-.593	.209	-.791	.214	-.756	.201	-.718	.209	-.666
-.023	-.463	.286	-.503	.294	-.730	.295	-.692	.294	-.664	.293	-.648
.133	-.394	.396	-.516	.404	-.625	.407	-.684	.397	-.626	.494	-.616
.272	-.345	.514	-.377	.497	-.484	.502	-.668	.495	-.618	.590	-.586
.416	-.322	.618	-.335	.599	-.317	.601	-.355	.594	-.627	.693	-.028
.565	-.213	.733	-.323	.700	-.435	.698	-.350	.693	-.327	.777	-.024
.713	-.135	.835	-.277	.864	-.165	.863	-.231	.784	-.325	.861	-.059
.854	-.172	.919	-.106	.926	-.076	.923	-.122	.856	-.194	.918	-.049
.980	-.217	.987	-.028	.975	-.035	.977	-.022	.926	-.096	.972	.048
1.074	-.241							.977	-.003		
1.122	-.136										
					LOWER SURFACE						
-.660	.068	-.022	.261	.024	.176	.025	.120	.019	.079	.020	.156
-.616	.055	.038	.046	.075	-.039	.130	-.097	.066	-.117	.076	-.125
-.572	.022	.101	-.043	.297	-.142	.298	-.180	.136	-.128	.136	-.209
-.462	-.015	.185	-.096	.400	-.113	.397	-.094	.214	-.131	.221	-.176
-.329	-.056	.398	-.123	.604	-.046	.501	-.168	.292	-.136	.295	-.188
-.172	-.121	.737	.117	.785	.193	.603	.032	.403	-.050	.396	-.174
-.030	-.205			.967	.223	.703	.077	.489	-.093	.497	-.176
.128	-.255			1.000	-.017	.784	.167	.594	-.158	.597	-.042
.418	-.156					.868	.252	.700	.085	.702	.109
.564	-.060					.923	.300	.786	.194	.786	.194
.710	.053					.972	.218	.858	.265	.864	.229
.976	.236							.919	.291	.912	.251
1.072	.218							.967	.176		
1.110	.152										
CN=	.4365		.4632		.5065		.5151		.4958		.3939
CM=	-.0238		-.0597		-.0825		-.0960		-.1012		-.0647

$\alpha = 3.97^{\circ}$; $C_L = 0.501$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.058	-.021	-.736	.023	-.732	.025	-.688	.022	-.562	.018	-.517
-.567	-.215	.035	-.868	.068	-.870	.079	-.780	.075	-.762	.077	-.708
-.452	-.291	.105	-.820	.134	-.841	.133	-.809	.129	-.739	.129	-.742
-.311	-.379	.178	-.621	.209	-.832	.214	-.786	.201	-.754	.209	-.715
-.023	-.464	.286	-.509	.294	-.775	.295	-.745	.294	-.705	.293	-.680
.133	-.405	.396	-.529	.404	-.718	.407	-.731	.397	-.697	.494	-.653
.272	-.354	.514	-.393	.497	-.522	.502	-.721	.495	-.666	.590	-.622
.416	-.324	.618	-.342	.599	-.348	.601	-.706	.594	-.664	.693	-.148
.565	-.207	.733	-.321	.700	-.437	.698	-.347	.693	-.672	.777	-.058
.713	-.137	.835	-.270	.864	-.182	.863	-.211	.784	-.272	.861	-.011
.854	-.180	.919	-.106	.926	-.085	.923	-.104	.856	-.172	.918	.029
.980	-.220	.987	-.028	.975	-.042	.977	-.052	.926	-.104	.972	.031
1.074	-.252							.977	-.068		
1.122	-.141										
					LOWER SURFACE						
-.660	.071	-.022	.307	.024	.236	.025	.160	.019	.126	.020	.179
-.616	.062	.038	.080	.075	.010	.130	-.075	.066	-.081	.076	-.109
-.572	.038	.101	-.004	.297	-.114	.298	-.162	.136	-.099	.136	-.188
-.462	-.007	.185	-.057	.400	-.094	.397	-.105	.214	-.118	.221	-.159
-.329	-.050	.398	-.094	.604	-.050	.501	-.162	.292	-.133	.295	-.175
-.172	-.115	.737	.126	.785	.189	.603	.031	.403	-.053	.396	-.188
-.030	-.183			.967	.210	.703	.073	.489	-.095	.497	-.201
.128	-.244			1.000	-.038	.784	.163	.594	-.174	.597	-.094
.418	-.100					.868	.250	.700	.069	.702	.056
.564	-.045					.923	.300	.786	.179	.786	.151
.710	.073					.972	.200	.858	.251	.864	.207
.976	.240							.919	.273	.912	.204
1.072	.217							.967	.151		
1.110	.154										
CN=	.4791		.5063		.5562		.5802		.5550		.4086
CM=	-.0212		-.0591		-.0835		-.1058		-.1120		-.0563

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(g) M = 0.98. Concluded.

$\alpha = 4.95^o$; $C_L = 0.601$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.115	-.021	-.841	.023	-.811	.025	-.781	.022	-.679	.018	-.625
	-.567	-.254	.035	-.996	.068	-.927	.079	-.857	.075	-.829	.077	-.786
	-.452	-.309	.105	-.924	.134	-.915	.133	-.872	.129	-.825	.129	-.811
	-.311	-.393	.178	-.845	.209	-.903	.214	-.867	.201	-.833	.209	-.796
	-.023	-.486	.286	-.533	.294	-.860	.295	-.816	.294	-.779	.293	-.757
	.133	-.408	.396	-.557	.404	-.865	.407	-.794	.397	-.772	.494	-.744
	.272	-.362	.514	-.427	.497	-.837	.502	-.798	.495	-.749	.590	-.291
	.416	-.324	.618	-.370	.599	-.463	.601	-.786	.594	-.753	.693	-.195
	.565	-.204	.733	-.340	.700	-.462	.698	-.577	.693	-.310	.777	-.152
	.713	-.140	.835	-.275	.864	-.170	.863	-.286	.784	-.273	.861	-.135
	.854	-.191	.919	-.119	.926	-.096	.923	-.228	.856	-.261	.918	-.120
	.980	-.243	.987	-.055	.975	-.083	.977	-.243	.926	-.267	.972	-.100
	1.074	-.292							.977	-.261		
	1.122	-.181										
LOWER SURFACE												
	-.660	.077	-.022	.363	.024	.318	.025	.251	.019	.203	.020	.243
	-.616	.084	.038	.154	.075	.095	.130	-.012	.066	-.014	.076	-.057
	-.572	.055	.101	.068	.297	-.046	.298	-.054	.136	-.057	.136	-.144
	-.462	.017	.185	-.001	.400	-.076	.397	-.091	.214	-.086	.221	-.185
	-.329	-.023	.398	-.056	.604	-.051	.501	-.129	.292	-.096	.295	-.166
	-.172	-.090	.737	.136	.785	.194	.603	.028	.403	-.040	.396	-.192
	-.030	-.132			.967	.197	.703	.062	.489	-.096	.497	-.240
	.128	-.222			1.000	-.059	.784	.145	.594	-.217	.597	-.174
	.418	-.059					.868	.241	.700	.027	.702	-.008
	.564	-.012					.923	.293	.786	.135	.786	.083
	.710	.091					.972	.161	.858	.216	.864	.134
	.976	.259							.919	.225	.912	.135
	1.072	.222							.967	.069		
	1.110	.154										
CN=	.5530		.6040		.6723		.7131		.5949		.4287	
CM=	-.0138		-.0600		-.0955		-.1353		-.1043		-.0492	

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(h) M = 0.99.

$\alpha = -4.96^o$; $C_L = -0.572$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE											
-.660	.055	-.021	.380	.023	.375	.025	.351	.022	.425	.018	.471
-.567	.031	.035	.220	.068	.228	.079	.221	.075	.240	.077	.237
-.452	-.074	.105	.151	.209	.086	.133	.169	.129	.178	.129	.157
-.311	-.173	.178	.113	.294	.005	.214	.114	.201	.114	.209	.086
-.023	-.261	.286	.054	.404	.031	.295	.094	.294	.074	.293	.038
.133	-.007	.396	.033	.497	.001	.407	.039	.397	.028	.494	-.123
.272	.061	.514	.031	.599	-.038	.502	-.014	.495	-.034	.590	-.204
.416	.091	.618	-.007	.700	-.069	.601	-.068	.594	-.107	.693	-.298
.565	.117	.733	-.043	.864	-.174	.698	-.150	.693	-.195	.777	-.382
.713	.107	.835	-.103	.926	-.217	.863	-.387	.784	-.307	.861	-.504
.854	.052	.919	-.071			.923	-.357	.856	-.414	.918	-.627
.980	.003	.987	.059			.977	-.187	.926	-.604	.972	-.423
1.074	-.073							.977	-.433		
LOWER SURFACE											
-.660	-.025	-.022	-.537	.024	-.891	.074	-1.081	.019	-1.020	.020	-.590
-.616	-.108	.038	-.836	.075	-.922	.130	-1.054	.066	-1.063	.076	-.625
-.462	-.206	.101	-.757	.297	-.934	.298	-.997	.136	-1.042	.136	-.545
-.329	-.213	.185	-.629	.400	-.902	.397	-.930	.214	-.985	.221	-.488
-.172	-.239	.398	-.542	.604	-.636	.501	-.530	.292	-.868	.295	-.450
-.030	-.307	.737	-.197	.785	.102	.603	-.302	.403	-.728	.396	-.438
.128	-.417			.967	.071	.784	.093	.489	-.673	.497	-.391
.418	-.286			1.000	.005	.868	.093	.594	-.648	.597	-.436
.564	-.306					.923	-.041	.700	-.518	.702	-.414
.710	-.265					.972	-.099	.786	-.417	.786	-.392
.976	.047							.858	-.366	.864	-.340
1.072	.118							.919	-.336	.912	-.355
1.110	.099							.967	-.330	.985	-.335
CN=	-.3460		-.5452		-.5545		-.5069		-.6099		-.3080
CM=	-.0047		.0123		-.0146		-.0546		.0091		-.0213

$\alpha = -3.92^o$; $C_L = -0.488$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE											
-.660	.050	-.021	.302	.023	.311	.025	.287	.022	.384	.018	.450
-.567	.030	.035	.133	.068	.139	.079	.156	.075	.196	.077	.212
-.452	-.095	.105	.077	.209	.039	.133	.106	.129	.134	.129	.137
-.311	-.195	.178	.031	.294	-.047	.214	.062	.201	.084	.209	.076
-.023	-.288	.286	.007	.404	-.011	.295	.052	.294	.045	.293	.017
.133	-.183	.396	-.015	.497	-.051	.407	.008	.397	.017	.494	-.113
.272	.015	.514	-.013	.599	-.065	.502	-.036	.495	-.047	.590	-.188
.416	.065	.618	-.052	.700	-.121	.601	-.081	.594	-.067	.693	-.286
.565	.089	.733	-.075	.864	-.154	.698	-.163	.693	-.136	.777	-.387
.713	.082	.835	-.147	.926	-.138	.863	-.321	.784	-.248	.861	-.486
.854	.029	.919	-.084			.923	-.119	.856	-.374	.918	-.635
.980	-.021	.987	.045			.977	-.014	.926	-.506	.972	-.365
1.074	-.090							.977	-.512		
LOWER SURFACE											
-.660	-.002	-.022	-.445	.024	-.786	.074	-1.017	.019	-.943	.020	-.559
-.616	-.087	.038	-.715	.075	-.826	.130	-.992	.066	-1.034	.076	-.539
-.462	-.176	.101	-.589	.297	-.854	.298	-.954	.136	-.979	.136	-.478
-.329	-.191	.185	-.561	.400	-.829	.397	-.934	.214	-.955	.221	-.448
-.172	-.221	.398	-.558	.604	-.264	.501	-.752	.292	-.870	.295	-.432
-.030	-.292	.737	-.151	.785	-.004	.603	-.195	.403	-.678	.396	-.387
.128	-.405			.967	.076	.784	.166	.489	-.615	.497	-.345
.418	-.329			1.000	-.020	.868	.147	.594	-.594	.597	-.358
.564	-.311					.923	.024	.700	-.492	.702	-.337
.710	-.233					.972	-.028	.786	-.297	.786	-.329
.976	.059							.858	-.219	.864	-.317
1.072	.121							.919	-.125	.912	-.305
1.110	.105							.967	-.138	.985	-.293
CN=	-.2623		-.4402		-.4267		-.4769		-.5422		-.2555
CM=	.0022		.0027		-.0267		-.0454		-.0108		-.0302

[redacted]

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = -2.97^{\circ}$; $C_L = -0.392$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	.043	-.021	.208	.023	.202	.025	.220	.022	.335	.018	.416
	-.567	.008	.035	.068	.068	.067	.079	.076	.075	.138	.077	.170
	-.452	-.112	.105	.003	.209	-.023	.133	.041	.129	.092	.129	.106
	-.311	-.225	.178	-.032	.294	-.092	.214	.008	.201	.043	.209	.048
	-.023	-.328	.286	-.060	.404	-.057	.295	.019	.294	.010	.293	-.003
	.133	-.232	.396	-.060	.497	-.101	.407	-.027	.397	-.012	.494	-.126
	.272	-.130	.514	-.053	.599	-.126	.502	-.074	.495	-.079	.590	-.197
	.416	.020	.618	-.075	.700	-.176	.601	-.107	.594	-.108	.693	-.287
	.565	.066	.733	-.107	.864	-.163	.698	-.186	.693	-.124	.777	-.386
	.713	.054	.835	-.163	.926	-.133	.863	-.249	.784	-.227	.861	-.477
	.854	.006	.919	-.100			.923	-.117	.856	-.350	.918	-.627
	.980	-.040	.987	.040			.977	.028	.926	-.438	.972	-.314
	1.074	-.101							.977	-.319		
LOWER SURFACE												
	-.660	-.003	-.022	-.323	.024	-.701	.074	-.946	.019	-.868	.020	-.463
	-.616	-.060	.038	-.528	.075	-.740	.130	-.924	.066	-.919	.076	-.527
	-.462	-.151	.101	-.530	.297	-.774	.298	-.897	.136	-.894	.136	-.442
	-.329	-.183	.185	-.521	.400	-.745	.397	-.845	.214	-.823	.221	-.364
	-.172	-.203	.398	-.513	.604	-.230	.501	-.640	.292	-.659	.295	-.392
	-.030	-.279	.737	-.116	.785	.019	.603	-.261	.403	-.529	.396	-.337
	.128	-.399			.967	.094	.784	.257	.489	-.502	.497	-.312
	.418	-.340			1.000	.026	.868	.201	.594	-.485	.597	-.287
	.564	-.327					.923	.060	.700	-.371	.702	-.283
	.710	-.273					.972	.051	.786	-.251	.786	-.271
	.976	.072							.858	-.181	.864	-.248
	1.072	.122							.919	-.154	.912	-.233
	1.110	.103							.967	-.188	.985	-.231
CN=		-.1830		-.3374		-.3257		-.3999		-.4384		-.1898
CM=		.0102		-.0047		-.0377		-.0522		-.0122		-.0426

$\alpha = -2.06^{\circ}$; $C_L = -0.294$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	.049	-.021	.075	.023	.118	.025	.123	.022	.284	.018	.370
	-.567	-.005	.035	-.146	.068	-.020	.079	.003	.075	.082	.077	.119
	-.452	-.123	.105	-.064	.209	-.079	.133	-.033	.129	.037	.129	.067
	-.311	-.236	.178	-.086	.294	-.140	.214	-.053	.201	-.009	.209	.021
	-.023	-.332	.286	-.096	.404	-.088	.295	-.050	.294	-.027	.293	-.023
	.133	-.247	.396	-.103	.497	-.132	.407	-.032	.397	-.041	.494	-.136
	.272	-.198	.514	-.086	.599	-.162	.502	-.083	.495	-.099	.590	-.192
	.416	-.054	.618	-.112	.700	-.231	.601	-.129	.594	-.165	.693	-.277
	.565	.033	.733	-.124	.864	-.143	.698	-.203	.693	-.154	.777	-.357
	.713	.034	.835	-.182	.926	-.100	.863	-.279	.784	-.208	.861	-.329
	.854	-.008	.919	-.106			.923	-.096	.856	-.279	.918	-.362
	.980	-.063	.987	.045			.977	.036	.926	-.193	.972	-.190
	1.074	-.103							.977	.033		
LOWER SURFACE												
	-.660	.016	-.022	-.168	.024	-.560	.074	-.868	.019	-.762	.020	-.701
	-.616	-.045	.038	-.470	.075	-.610	.130	-.860	.066	-.932	.076	-.654
	-.462	-.127	.101	-.457	.297	-.679	.298	-.822	.136	-.903	.136	-.542
	-.329	-.164	.185	-.490	.400	-.657	.397	-.806	.214	-.892	.221	-.473
	-.172	-.195	.398	-.478	.604	-.217	.501	-.548	.292	-.810	.295	-.393
	-.030	-.264	.737	-.083	.785	.049	.603	-.099	.403	-.542	.396	-.356
	.128	-.383			.967	.136	.784	.249	.489	-.464	.497	-.274
	.418	-.320			1.000	.053	.868	.284	.594	-.403	.597	-.265
	.564	-.325					.923	.206	.700	-.289	.702	-.234
	.710	-.220					.972	.149	.786	-.005	.786	-.234
	.976	.084							.858	.146	.864	-.201
	1.072	.134							.919	.195	.912	-.182
	1.110	.105							.967	.177	.985	-.161
CN=		-.1183		-.2407		-.2253		-.2867		-.3632		-.2246
CM=		.0103		-.0090		-.0429		-.0745		-.0458		-.0354

 [redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = -1.14^{\circ}$; $C_L = -0.179$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.036	-.021	-.025	.023	-.011	.025	.049	.022	.220	.018	.335
-.567	-.037	.035	-.232	.068	-.156	.079	-.083	.075	.036	.077	.071
-.452	-.147	.105	-.244	.209	-.169	.133	-.090	.129	-.005	.129	.011
-.311	-.254	.178	-.276	.294	-.158	.214	-.101	.201	-.040	.209	-.022
-.023	-.356	.286	-.095	.404	-.121	.295	-.102	.294	-.043	.293	-.059
.133	-.276	.396	-.146	.497	-.158	.407	-.106	.397	-.054	.494	-.137
.272	-.225	.514	-.134	.599	-.179	.502	-.058	.495	-.123	.590	-.210
.416	-.166	.618	-.157	.700	-.257	.601	-.117	.594	-.173	.693	-.306
.565	-.016	.733	-.150	.864	-.125	.698	-.202	.693	-.245	.777	-.405
.713	.009	.835	-.189	.926	-.054	.863	-.350	.784	-.218	.861	-.155
.854	-.039	.919	-.091			.923	-.087	.856	-.308	.918	-.017
.980	-.082	.987	.045			.977	.045	.926	-.200	.972	.025
1.074	-.117							.977	-.002		
				LOWER SURFACE							
-.660	.024	-.022	-.078	.024	-.383	.074	-.781	.019	-.669	.020	-.730
-.616	-.022	.038	-.368	.075	-.521	.130	-.787	.066	-.879	.076	-.654
-.462	-.107	.101	-.407	.297	-.571	.298	-.733	.136	-.841	.136	-.630
-.329	-.143	.185	-.445	.400	-.568	.397	-.720	.214	-.837	.221	-.539
-.172	-.178	.398	-.452	.604	-.167	.501	-.162	.292	-.781	.295	-.404
-.030	-.257	.737	-.045	.785	.133	.603	.031	.403	-.371	.396	-.354
.128	-.368			.967	.169	.784	.194	.489	-.308	.497	-.278
.418	-.301			1.000	.082	.868	.273	.594	-.137	.597	-.221
.564	-.320					.923	.242	.700	.002	.702	-.178
.710	-.151					.972	.187	.786	.126	.786	-.098
.976	.103							.858	.173	.864	-.075
1.072	.139							.919	.219	.912	-.037
1.110	.119							.967	.146	.985	-.015
CN=	-.0266		-.1365		-.1093		-.1622		-.2144		-.2120
CM=	.0076		-.0138		-.0512		-.0912		-.0850		-.0364

$\alpha = 0.02^{\circ}$; $C_L = -0.024$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.017	-.021	-.146	.023	-.265	.025	-.106	.022	.128	.018	.236
-.567	-.063	.035	-.400	.068	-.384	.079	-.227	.075	-.036	.077	-.015
-.452	-.175	.105	-.316	.209	-.367	.133	-.196	.129	-.055	.129	-.048
-.311	-.282	.178	-.340	.294	-.296	.214	-.186	.201	-.074	.209	-.080
-.023	-.382	.286	-.356	.404	-.177	.295	-.177	.294	-.071	.293	-.112
.133	-.310	.396	-.273	.497	-.203	.407	-.136	.397	-.084	.494	-.213
.272	-.262	.514	-.206	.599	-.273	.502	-.034	.495	-.137	.590	-.256
.416	-.212	.618	-.217	.700	-.298	.601	-.107	.594	-.203	.693	-.346
.565	-.095	.733	-.213	.864	-.088	.698	-.209	.693	-.291	.777	-.436
.713	-.052	.835	-.250	.926	-.010	.863	-.356	.784	-.357	.861	-.157
.854	-.103	.919	-.098			.923	-.117	.856	-.314	.918	.005
.980	-.138	.987	.045			.977	.020	.926	-.181	.972	.068
1.074	-.166							.977	-.019		
				LOWER SURFACE							
-.660	.044	-.022	.022	.024	-.246	.074	-.659	.019	-.548	.020	-.558
-.616	-.003	.038	-.262	.075	-.426	.130	-.654	.066	-.718	.076	-.697
-.462	-.084	.101	-.338	.297	-.489	.298	-.590	.136	-.635	.136	-.709
-.329	-.131	.185	-.335	.400	-.540	.397	-.178	.214	-.363	.221	-.667
-.172	-.159	.398	-.409	.604	-.135	.501	-.030	.292	-.231	.295	-.346
-.030	-.241	.737	-.002	.785	.174	.603	.073	.403	-.200	.396	-.324
.128	-.335			.967	.233	.784	.190	.489	-.194	.497	-.197
.418	-.276			1.000	.079	.868	.230	.594	-.165	.597	-.006
.564	-.281					.923	.257	.700	.007	.702	.094
.710	-.149					.972	.199	.786	.113	.786	.138
.976	.123							.858	.186	.864	.166
1.072	.151							.919	.205	.912	.185
1.110	.122							.967	.127	.985	.082
CN=	.0950		.0304		.0343		-.0014		-.0323		-.0605
CM=	-.0060		-.0295		-.0558		-.0992		-.0919		-.0927

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(h) M = 0.99. Continued.

$\alpha = 1.03^o$; $C_L = 0.104$

	STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE												
	-.660	.003	-.021	-.329	.023	-.385	.025	-.319	.022	-.038	.018	.095
	-.567	-.095	.035	-.601	.069	-.499	.079	-.444	.075	-.153	.077	-.140
	-.452	-.204	.105	-.439	.209	-.483	.133	-.440	.129	-.112	.129	-.143
	-.311	-.303	.178	-.388	.294	-.500	.214	-.373	.201	-.126	.209	-.149
	-.023	-.409	.286	-.395	.404	-.255	.295	-.333	.294	-.110	.293	-.176
	.133	-.324	.396	-.384	.497	-.225	.407	-.214	.397	-.105	.494	-.260
	.272	-.281	.514	-.258	.599	-.288	.502	-.253	.495	-.149	.590	-.307
	.416	-.250	.618	-.254	.700	-.347	.601	-.036	.594	-.216	.693	-.400
	.565	-.143	.733	-.252	.864	-.041	.698	-.158	.693	-.306	.777	-.501
	.713	-.079	.835	-.277	.926	.034	.863	-.339	.784	-.393	.861	-.098
	.854	-.128	.919	-.122			.923	-.170	.856	-.503	.918	.007
	.980	-.171	.987	.020			.977	.011	.926	-.156	.972	.058
	1.074	-.213							.977	-.039		
LOWER SURFACE												
	-.660	.051	-.022	.098	.024	-.124	.074	-.566	.019	-.260	.020	-.308
	-.616	.010	.038	-.196	.075	-.349	.130	-.538	.066	-.384	.076	-.541
	-.462	-.066	.101	-.282	.297	-.435	.298	-.428	.136	-.308	.136	-.499
	-.329	-.105	.135	-.281	.400	-.444	.397	.013	.214	-.234	.221	-.412
	-.172	-.146	.398	-.371	.604	-.097	.501	-.043	.292	-.226	.295	-.268
	-.030	-.232	.737	.034	.785	.209	.603	.040	.403	-.260	.396	-.370
	.128	-.314			.967	.274	.784	.188	.489	-.121	.497	-.265
	.418	-.257			1.000	.079	.868	.248	.594	-.206	.597	-.020
	.564	-.248					.923	.272	.700	.048	.702	.104
	.710	-.110					.972	.207	.786	.152	.786	.168
	.976	.142							.858	.214	.864	.213
	1.072	.161							.919	.237	.912	.216
	1.110	.119							.967	.156	.985	.088
CN=	.1843		.1504		.1499		.1449		.0912		.0520	
CM=	-.0115		-.0398		-.0608		-.0954		-.0963		-.0915	

$\alpha = 1.95^o$; $C_L = 0.226$

	STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE												
	-.660	-.006	-.021	-.432	.023	-.482	.025	-.437	.022	-.341	.018	-.046
	-.567	-.138	.035	-.667	.068	-.647	.079	-.554	.075	-.523	.077	-.222
	-.452	-.223	.105	-.517	.209	-.528	.133	-.570	.129	-.492	.129	-.202
	-.311	-.318	.178	-.468	.294	-.551	.214	-.518	.201	-.409	.209	-.161
	-.023	-.414	.286	-.447	.404	-.457	.295	-.481	.294	-.388	.293	-.179
	.133	-.343	.396	-.455	.497	-.289	.407	-.327	.397	-.139	.494	-.253
	.272	-.295	.514	-.309	.599	-.298	.502	-.232	.495	.029	.590	-.296
	.416	-.276	.618	-.295	.700	-.384	.601	-.290	.594	-.058	.693	-.366
	.565	-.177	.733	-.284	.864	-.103	.698	-.138	.693	-.263	.777	-.476
	.713	-.110	.835	-.303	.926	-.013	.863	-.167	.784	-.332	.861	-.124
	.854	-.146	.919	-.138			.923	-.133	.856	-.481	.918	-.004
	.980	-.189	.987	-.003			.977	.038	.926	-.251	.972	.033
	1.074	-.233							.977	-.054		
LOWER SURFACE												
	-.660	.063	-.022	.137	.024	-.041	.074	-.224	.019	-.116	.020	-.005
	-.616	.030	.038	-.133	.075	-.269	.130	-.172	.066	-.331	.076	-.257
	-.462	-.040	.101	-.176	.297	-.367	.298	-.196	.136	-.257	.136	-.283
	-.329	-.078	.185	-.237	.400	-.355	.397	-.231	.214	-.134	.221	-.294
	-.172	-.134	.398	-.322	.604	-.119	.501	-.146	.292	-.130	.295	-.300
	-.030	-.217	.737	.085	.785	.197	.603	.055	.403	-.050	.396	-.217
	.128	-.290			.967	.241	.784	.195	.489	-.098	.497	-.228
	.418	-.238			1.000	.025	.868	.247	.594	-.138	.597	-.040
	.564	-.205					.923	.266	.700	.076	.702	.108
	.710	-.069					.972	.226	.786	.186	.786	.182
	.976	.183							.858	.244	.864	.222
	1.072	.178							.919	.274	.912	.231
	1.110	.129							.967	.154	.985	.059
CN=	.2681		.2629		.2514		.2760		.2356		.1366	
CM=	-.0162		-.0555		-.0664		-.0726		-.0855		-.0828	

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = 2.50^{\circ}$; $C_L = 0.303$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
				UPPER SURFACE							
-.660	-.025	-.021	-.555	.023	-.542	.025	-.505	.022	-.416	.018	-.322
-.567	-.158	.035	-.712	.068	-.695	.079	-.610	.075	-.588	.077	-.570
-.452	-.244	.105	-.634	.209	-.619	.133	-.638	.129	-.591	.129	-.566
-.311	-.337	.178	-.512	.294	-.582	.214	-.621	.201	-.595	.209	-.486
-.023	-.432	.286	-.469	.404	-.467	.295	-.556	.294	-.549	.293	-.460
.133	-.360	.396	-.465	.497	-.434	.407	-.475	.397	-.547	.494	-.067
.272	-.308	.514	-.354	.599	-.307	.502	-.327	.495	-.232	.590	-.126
.416	-.288	.618	-.316	.700	-.402	.601	-.303	.594	-.227	.693	-.275
.565	-.198	.733	-.305	.864	-.140	.698	-.387	.693	-.133	.777	-.397
.713	-.124	.835	-.311	.926	-.053	.863	-.145	.784	-.239	.861	-.134
.854	-.169	.919	-.140			.923	-.061	.856	-.320	.918	-.021
.980	-.208	.987	-.020			.977	.033	.926	-.203	.972	.054
1.074	-.253							.977	-.010		
				LOWER SURFACE							
-.660	.068	-.022	.163	.024	.028	.074	-.177	.019	-.034	.020	.085
-.616	.044	.038	-.067	.075	-.202	.130	-.194	.066	-.244	.076	-.183
-.462	-.032	.101	-.154	.297	-.200	.298	-.144	.136	-.199	.136	-.237
-.329	-.073	.185	-.210	.400	-.037	.397	-.198	.214	-.150	.221	-.245
-.172	-.128	.398	-.271	.604	-.147	.501	-.196	.292	-.165	.295	-.207
-.030	-.203	.737	.109	.785	.188	.603	.049	.403	-.064	.396	-.176
.128	-.273			.967	.236	.784	.182	.489	-.081	.497	-.115
.418	-.218			1.000	.008	.868	.255	.594	-.143	.597	.017
.564	-.179					.923	.291	.700	.099	.702	.121
.710	-.026					.972	.236	.786	.214	.786	.210
.976	.210							.858	.277	.864	.264
1.072	.198							.919	.304	.912	.266
1.110	.145							.967	.203	.985	.094
CN=	.3351		.3358		.3794		.3621		.3459		.2560
CM=	-.0230		-.0620		-.0782		-.0859		-.0856		-.0642

$\alpha = 2.91^{\circ}$; $C_L = 0.360$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
				UPPER SURFACE							
-.660	-.027	-.021	-.585	.023	-.571	.025	-.562	.022	-.430	.018	-.358
-.567	-.167	.035	-.735	.068	-.727	.079	-.660	.075	-.627	.077	-.599
-.452	-.249	.105	-.684	.209	-.690	.133	-.668	.129	-.619	.129	-.613
-.311	-.336	.178	-.511	.294	-.608	.214	-.664	.201	-.638	.209	-.591
-.023	-.440	.286	-.477	.404	-.473	.295	-.616	.294	-.581	.293	-.547
.133	-.367	.396	-.486	.497	-.462	.407	-.576	.397	-.571	.494	-.369
.272	-.317	.514	-.383	.599	-.313	.502	-.482	.495	-.570	.590	-.104
.416	-.295	.618	-.331	.700	-.423	.601	-.315	.594	-.310	.693	-.028
.565	-.208	.733	-.312	.864	-.194	.698	-.384	.693	-.290	.777	-.056
.713	-.139	.835	-.310	.926	-.064	.863	-.144	.784	-.177	.861	-.085
.854	-.177	.919	-.136			.923	-.052	.856	-.256	.918	-.089
.980	-.213	.987	-.037			.977	.017	.926	-.137	.972	.060
1.074	-.256							.977	.015		
				LOWER SURFACE							
-.660	.073	-.022	.196	.024	.106	.074	-.157	.019	.006	.020	.096
-.616	.045	.038	-.055	.075	-.140	.130	-.186	.066	-.196	.076	-.157
-.462	-.018	.101	-.135	.297	-.155	.298	-.196	.136	-.158	.136	-.237
-.329	-.066	.185	-.181	.400	-.172	.397	-.243	.214	-.157	.221	-.235
-.172	-.121	.398	-.206	.604	-.130	.501	-.185	.292	-.167	.295	-.213
-.030	-.202	.737	.104	.785	.182	.603	.041	.403	-.042	.396	-.172
.128	-.265			.967	.217	.784	.169	.489	-.079	.497	-.153
.418	-.214			1.000	-.030	.868	.245	.594	-.167	.597	-.025
.564	-.143					.923	.287	.700	.089	.702	.130
.710	.004					.972	.217	.786	.205	.786	.217
.976	.218							.858	.273	.864	.255
1.072	.210							.919	.297	.912	.266
1.110	.143							.967	.197	.985	.144
CN=	.3686		.3772		.4078		.3934		.4053		.2723
CM=	-.0274		-.0640		-.0779		-.0868		-.0906		-.0478

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = 3.46^{\circ}$; $C_L = 0.430$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.046	-.021	-.673	.023	-.640	.025	-.606	.022	-.484	.018	-.414
-.567	-.183	.035	-.775	.068	-.777	.079	-.699	.075	-.682	.077	-.656
-.452	-.266	.105	-.764	.209	-.744	.133	-.720	.129	-.673	.129	-.666
-.311	-.352	.178	-.560	.294	-.708	.214	-.722	.201	-.694	.209	-.651
-.023	-.452	.286	-.496	.404	-.645	.295	-.676	.294	-.647	.293	-.599
.133	-.380	.396	-.502	.497	-.476	.407	-.652	.397	-.636	.494	-.598
.272	-.326	.514	-.395	.599	-.352	.502	-.646	.495	-.624	.590	-.590
.416	-.307	.618	-.350	.700	-.445	.601	-.464	.594	-.615	.693	-.161
.565	-.220	.733	-.327	.864	-.237	.698	-.407	.693	-.578	.777	-.038
.713	-.147	.835	-.315	.926	-.085	.863	-.151	.784	-.221	.861	.003
.854	-.189	.919	-.141			.923	-.069	.856	-.148	.918	.025
.980	-.223	.987	-.050			.977	-.033	.926	-.068	.972	.079
1.074	-.270							.977	-.021		
					LOWER SURFACE						
-.660	.076	-.022	.240	.024	.180	.074	-.099	.019	.046	.020	.142
-.616	.059	.038	-.003	.075	-.063	.130	-.103	.066	-.136	.076	-.136
-.462	-.010	.101	-.096	.297	-.131	.298	-.180	.136	-.149	.136	-.211
-.329	-.049	.185	-.152	.400	-.133	.397	-.228	.214	-.133	.221	-.228
-.172	-.110	.398	-.132	.604	-.096	.501	-.161	.292	-.158	.295	-.203
-.030	-.187	.737	.116	.785	.177	.603	.035	.403	-.139	.396	-.179
.128	-.250			.967	.208	.784	.159	.489	-.081	.497	-.181
.418	-.193			1.000	-.050	.868	.242	.594	-.177	.597	-.062
.564	-.102					.923	.290	.700	.078	.702	.090
.710	.049					.972	.206	.786	.188	.786	.172
.976	.230							.858	.259	.864	.228
1.072	.218							.919	.279	.912	.222
1.110	.146							.967	.172	.985	.130
CN=	.4264		.4440		.4928		.4757		.4801		.3607
CM=	-.0291		-.0680		-.0865		-.0976		-.1031		-.0570

$\alpha = 3.94^{\circ}$; $C_L = 0.484$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.067	-.021	-.734	.023	-.687	.025	-.655	.022	-.538	.018	-.479
-.567	-.199	.035	-.848	.068	-.818	.079	-.748	.075	-.727	.077	-.696
-.452	-.278	.105	-.808	.209	-.804	.133	-.770	.129	-.718	.129	-.708
-.311	-.367	.178	-.605	.294	-.775	.214	-.753	.201	-.728	.209	-.677
-.023	-.467	.286	-.506	.404	-.729	.295	-.719	.294	-.700	.293	-.668
.133	-.384	.396	-.521	.497	-.497	.407	-.698	.397	-.690	.494	-.647
.272	-.340	.514	-.419	.599	-.388	.502	-.687	.495	-.666	.590	-.633
.416	-.312	.618	-.362	.700	-.457	.601	-.699	.594	-.667	.693	-.180
.565	-.227	.733	-.340	.864	-.243	.698	-.555	.693	-.452	.777	-.095
.713	-.156	.835	-.313	.926	-.104	.863	-.154	.784	-.218	.861	-.045
.854	-.190	.919	-.146			.923	-.097	.856	-.176	.918	-.011
.980	-.232	.987	-.058			.977	-.070	.926	-.146	.972	.014
1.074	-.282							.977	-.114		
					LOWER SURFACE						
-.660	.079	-.022	.283	.024	.224	.074	-.031	.019	.100	.020	.170
-.616	.064	.038	.035	.075	-.007	.130	-.081	.066	-.099	.076	-.103
-.462	.001	.101	-.034	.297	-.126	.298	-.168	.136	-.114	.136	-.193
-.329	-.040	.185	-.058	.400	-.118	.397	-.168	.214	-.120	.221	-.209
-.172	-.100	.398	-.108	.604	-.041	.501	-.153	.292	-.145	.295	-.234
-.030	-.178	.737	.121	.785	.176	.603	.032	.403	-.028	.396	-.177
.128	-.234			.967	.200	.784	.152	.489	-.086	.497	-.226
.418	-.161			1.000	-.061	.868	.237	.594	-.197	.597	-.112
.564	-.054					.923	.293	.700	.059	.702	.037
.710	.065					.972	.187	.786	.165	.786	.129
.976	.241							.858	.236	.864	.181
1.072	.221							.919	.256	.912	.188
1.110	.146							.967	.112	.985	.067
CN=	.4751		.5052		.5530		.5616		.5174		.3849
CM=	-.0283		-.0674		-.0926		-.1157		-.1012		-.0550

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; δ_h = -2.5°

(h) M = 0.99. Continued.

α = 4.92°; C_L = 0.592

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.113	-.021	-.824	.023	-.776	.025	-.758	.022	-.647	.018	-.571
	-.567	-.242	.035	-.949	.068	-.916	.079	-.831	.075	-.802	.077	-.769
	-.452	-.309	.105	-.912	.209	-.870	.133	-.839	.129	-.796	.129	-.792
	-.311	-.387	.178	-.851	.294	-.863	.214	-.834	.201	-.806	.209	-.767
	-.023	-.482	.286	-.533	.404	-.840	.295	-.800	.294	-.773	.293	-.736
	.133	-.402	.396	-.544	.497	-.810	.407	-.793	.397	-.760	.494	-.725
	.272	-.355	.514	-.452	.599	-.524	.502	-.774	.495	-.744	.590	-.717
	.416	-.324	.618	-.391	.700	-.482	.601	-.774	.594	-.728	.693	-.445
	.565	-.224	.733	-.358	.864	-.256	.698	-.789	.693	-.367	.777	-.237
	.713	-.163	.835	-.320	.926	-.127	.863	-.342	.784	-.308	.861	-.198
	.854	-.200	.919	-.153			.923	-.318	.856	-.257	.918	-.180
	.980	-.245	.987	-.081			.977	-.297	.926	-.217	.972	-.160
	1.074	-.309							.977	-.205		
LOWER SURFACE												
	-.660	.078	-.022	.353	.024	.309	.074	.038	.019	.195	.020	.227
	-.616	.080	.038	.153	.075	.083	.130	-.016	.066	-.025	.076	-.051
	-.462	.020	.101	.059	.297	-.072	.298	-.128	.136	-.069	.136	-.151
	-.329	-.020	.185	.009	.400	-.073	.397	-.082	.214	-.088	.221	-.184
	-.172	-.089	.398	-.062	.604	-.045	.501	-.173	.292	-.124	.295	-.224
	-.030	-.141	.737	.131	.785	.187	.603	.033	.403	-.011	.396	-.186
	.128	-.216			.967	.187	.784	.119	.489	-.083	.497	-.245
	.418	-.060			1.000	-.087	.868	.223	.594	-.217	.597	-.181
	.564	-.006					.923	.279	.700	.006	.702	-.026
	.710	.092					.972	.143	.786	.135	.786	.066
	.976	.263							.858	.207	.864	.120
	1.072	.234							.919	.221	.912	.118
	1.110	.149							.967	.056	.985	-.068
CN=	.5662		.6103		.6767		.7100		.5814		.4781	
CM=	-.0259		-.0672		-.1065		-.1569		-.1028		-.0751	

α = 5.88°; C_L = 0.680

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.153	-.021	-.915	.023	-.871	.025	-.833	.022	-.741	.018	-.669
	-.567	-.275	.035	-1.034	.068	-.962	.079	-.898	.075	-.871	.077	-.838
	-.452	-.334	.105	-1.012	.209	-.931	.133	-.910	.129	-.862	.129	-.846
	-.311	-.404	.178	-.980	.294	-.936	.214	-.887	.201	-.866	.209	-.830
	-.023	-.494	.286	-.613	.404	-.905	.295	-.852	.294	-.833	.293	-.796
	.133	-.413	.396	-.560	.497	-.884	.407	-.848	.397	-.817	.494	-.777
	.272	-.366	.514	-.468	.599	-.760	.502	-.836	.495	-.805	.590	-.763
	.416	-.330	.618	-.411	.700	-.556	.601	-.825	.594	-.765	.693	-.587
	.565	-.224	.733	-.374	.864	-.316	.698	-.567	.693	-.422	.777	-.318
	.713	-.164	.835	-.322	.926	-.176	.863	-.387	.784	-.362	.861	-.287
	.854	-.211	.919	-.157			.923	-.380	.856	-.287	.918	-.275
	.980	-.267	.987	-.086			.977	-.376	.926	-.255	.972	-.270
	1.074	-.325							.977	-.243		
LOWER SURFACE												
	-.660	.065	-.022	.411	.024	.380	.074	.100	.019	.272	.020	.285
	-.616	.097	.038	.212	.075	.151	.130	.038	.066	.045	.076	-.006
	-.462	.045	.101	.117	.297	-.023	.298	-.018	.136	-.021	.136	-.101
	-.329	.001	.185	.068	.400	-.050	.397	-.069	.214	-.054	.221	-.152
	-.172	-.059	.398	-.018	.604	-.037	.501	-.139	.292	-.092	.295	-.201
	-.030	-.096	.737	.149	.785	.202	.603	.033	.403	-.013	.396	-.253
	.128	-.194			.967	.194	.784	.102	.489	-.079	.497	-.237
	.418	-.018			1.000	-.100	.868	.215	.594	-.216	.597	-.199
	.564	.023					.923	.280	.700	-.006	.702	-.047
	.710	.120					.972	.120	.786	.122	.786	.037
	.976	.277							.858	.200	.864	.089
	1.072	.240							.919	.214	.912	.091
	1.110	.158							.967	.048	.985	-.146
CN=	.6373		.6984		.7881		.7598		.6492		.5478	
CM=	-.0178		-.0689		-.1291		-.1520		-.1093		-.0874	

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = 6.82^{\circ}$; $C_L = 0.782$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	-.176	-.021	-.964	.023	-.922	.025	-.903	.022	-.828	.018	-.736
-.567	-.335	.035	-1.098	.068	-1.034	.079	-.957	.075	-.932	.077	-.896
-.452	-.353	.105	-1.081	.209	-.990	.133	-.962	.129	-.917	.129	-.908
-.311	-.440	.178	-1.066	.294	-.983	.214	-.950	.201	-.925	.209	-.881
-.023	-.512	.286	-.823	.404	-.966	.295	-.899	.294	-.882	.293	-.849
.133	-.423	.396	-.599	.497	-.945	.407	-.904	.397	-.865	.494	-.835
.272	-.371	.514	-.491	.599	-.915	.502	-.897	.495	-.859	.590	-.817
.416	-.321	.618	-.436	.700	-.740	.601	-.887	.594	-.821	.693	-.813
.565	-.233	.733	-.388	.864	-.359	.698	-.461	.693	-.515	.777	-.383
.713	-.184	.835	-.323	.926	-.193	.863	-.424	.784	-.394	.861	-.336
.854	-.235	.919	-.168			.923	-.424	.856	-.353	.918	-.333
.980	-.282	.987	-.097			.977	-.422	.926	-.324	.972	-.336
1.074	-.346							.977	-.308		
				LOWER SURFACE							
-.660	.088	-.022	.447	.024	.434	.074	.167	.019	.330	.020	.324
-.616	.117	.033	.276	.075	.207	.130	.091	.066	.093	.076	.041
-.462	.072	.101	.187	.297	.028	.298	.012	.136	.023	.136	-.067
-.329	.073	.185	.117	.400	-.006	.397	-.045	.214	-.024	.221	-.125
-.172	-.028	.398	.027	.604	-.023	.501	-.092	.292	-.060	.295	-.180
-.030	-.067	.737	.164	.785	.212	.603	.037	.403	-.010	.396	-.170
.128	-.105			.967	.204	.784	.079	.489	-.083	.497	-.243
.418	.021			1.000	-.095	.868	.210	.594	-.212	.597	-.206
.564	.056					.923	.276	.700	-.012	.702	-.068
.710	.147					.972	.110	.786	.113	.786	.016
.976	.295							.858	.192	.864	.069
1.072	.251							.919	.211	.912	.069
1.110	.164							.967	.037	.985	-.198
CN=	.7343		.7932		.8981		.8132		.7206		.6299
CM=	-.0077		-.0736		-.1526		-.1522		-.1216		-.1030

$\alpha = 7.92^{\circ}$; $C_L = 0.875$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	-.227	-.021	-1.052	.023	-.996	.025	-.965	.022	-.899	.018	-.815
-.567	-.376	.035	-1.154	.068	-1.082	.079	-1.017	.075	-.995	.077	-.956
-.452	-.394	.105	-1.143	.209	-1.052	.133	-1.014	.129	-.975	.129	-.962
-.311	-.456	.178	-1.132	.294	-1.046	.214	-.997	.201	-.969	.209	-.934
-.023	-.523	.286	-.971	.404	-1.018	.295	-.957	.294	-.934	.293	-.900
.133	-.429	.396	-.772	.497	-.999	.407	-.955	.397	-.917	.494	-.890
.272	-.370	.514	-.591	.599	-.967	.502	-.956	.495	-.911	.590	-.876
.416	-.332	.618	-.530	.700	-.827	.601	-.935	.594	-.872	.693	-.866
.565	-.249	.733	-.407	.864	-.403	.698	-.522	.693	-.591	.777	-.486
.713	-.215	.835	-.362	.926	-.284	.863	-.477	.784	-.396	.861	-.426
.854	-.262	.919	-.211			.923	-.458	.856	-.363	.918	-.406
.980	-.305	.987	-.119			.977	-.455	.926	-.345	.972	-.399
1.074	-.365							.977	-.341		
				LOWER SURFACE							
-.660	.063	-.022	.472	.024	.489	.074	.220	.019	.365	.020	.365
-.616	.137	.038	.336	.075	.289	.130	.137	.066	.150	.076	.087
-.462	.093	.101	.247	.297	.073	.298	.035	.136	.060	.136	-.019
-.329	.061	.185	.168	.400	.025	.397	-.022	.214	.013	.221	-.090
-.172	.018	.398	.076	.604	-.015	.501	-.072	.292	-.024	.295	-.149
-.030	-.024	.737	.183	.785	.190	.603	.034	.403	-.004	.396	-.148
.128	-.043			.967	.132	.784	.084	.489	-.067	.497	-.229
.418	.057			1.000	-.324	.868	.226	.594	-.199	.597	-.208
.564	.095					.923	.288	.700	-.001	.702	-.066
.710	.181					.972	.123	.786	.116	.786	.014
.976	.312							.858	.197	.864	.069
1.072	.263							.919	.215	.912	.065
1.110	.174							.967	.023	.985	-.263
CN=	.8306		.9200		.9759		.8867		.7847		.7069
CM=	-.0003		-.0907		-.1606		-.1655		-.1295		-.1178

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(h) M = 0.99. Continued.

$\alpha = 9.08^o$; $C_L = 0.960$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.286	-.021	-1.106	.023	-1.055	.025	-1.024	.022	-.956	.018	-.886
-.567	-.447	.035	-1.220	.068	-1.146	.079	-1.069	.075	-1.054	.077	-1.019
-.452	-.430	.105	-1.176	.209	-1.078	.133	-1.072	.129	-1.032	.129	-1.016
-.311	-.477	.178	-1.162	.294	-1.070	.214	-1.054	.201	-1.034	.209	-.992
-.023	-.535	.286	-1.005	.404	-1.027	.295	-1.027	.294	-.995	.293	-.954
.133	-.436	.396	-.864	.497	-.874	.407	-1.016	.397	-.977	.494	-.939
.272	-.383	.514	-.772	.599	-.694	.502	-1.005	.495	-.968	.590	-.931
.416	-.346	.618	-.705	.700	-.597	.601	-.889	.594	-.942	.693	-.814
.565	-.278	.733	-.654	.864	-.471	.698	-.684	.693	-.894	.777	-.584
.713	-.241	.835	-.480	.926	-.441	.863	-.556	.784	-.609	.861	-.510
.854	-.282	.919	-.344			.923	-.501	.856	-.498	.918	-.471
.980	-.329	.987	-.181			.977	-.394	.926	-.466	.972	-.450
1.074	-.339							.977	-.473		
				LOWER SURFACE							
-.660	.079	-.022	.493	.024	.535	.074	.266	.019	.412	.020	.393
-.616	.152	.038	.386	.075	.336	.130	.171	.066	.196	.076	.119
-.462	.135	.101	.294	.297	.121	.298	.054	.136	.093	.136	.008
-.329	.092	.185	.222	.400	.063	.397	-.013	.214	.031	.221	-.058
-.172	.069	.398	.114	.604	-.000	.501	-.106	.292	-.011	.295	-.078
-.030	.078	.737	.197	.785	.175	.603	.024	.403	.001	.396	-.150
.128	.010			.967	.102	.784	.070	.489	-.064	.497	-.224
.418	.107			1.000	-.393	.868	.225	.594	-.193	.597	-.206
.564	.125					.923	.287	.700	.007	.702	-.074
.710	.207					.972	.132	.786	.117	.786	.005
.976	.341							.858	.183	.864	.058
1.072	.272							.919	.205	.912	.052
1.110	.189							.967	.012	.985	-.312
CN=	.9405		1.0720		.9634		.9495		.9033		.7674
CM=	.0107		-.1338		-.1454		-.1759		-.1681		-.1260

$\alpha = 10.04^o$; $C_L = 1.032$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.330	-.021	-1.139	.023	-1.113	.025	-1.078	.022	-1.020	.018	-.942
-.567	-.483	.035	-1.226	.068	-1.185	.079	-1.115	.075	-1.092	.077	-1.067
-.452	-.448	.105	-1.194	.209	-1.128	.133	-1.106	.129	-1.077	.129	-1.060
-.311	-.498	.178	-1.181	.294	-1.093	.214	-1.095	.201	-1.068	.209	-1.031
-.023	-.535	.286	-1.012	.404	-.944	.295	-1.056	.294	-1.036	.293	-1.013
.133	-.435	.396	-.954	.497	-.795	.407	-1.032	.397	-1.017	.494	-.971
.272	-.390	.514	-.867	.599	-.695	.502	-1.001	.495	-.986	.590	-.936
.416	-.353	.618	-.862	.700	-.603	.601	-.868	.594	-.959	.693	-.758
.565	-.291	.733	-.753	.864	-.489	.698	-.702	.693	-.887	.777	-.584
.713	-.258	.835	-.628	.926	-.471	.863	-.559	.784	-.757	.861	-.520
.854	-.306	.919	-.389			.923	-.472	.856	-.661	.918	-.491
.980	-.337	.987	-.197			.977	-.382	.926	-.606	.972	-.484
1.074	-.400							.977	-.595		
				LOWER SURFACE							
-.660	.073	-.022	.504	.024	.568	.074	.310	.019	.438	.020	.429
-.616	.175	.038	.427	.075	.375	.130	.219	.066	.239	.076	.160
-.462	.168	.101	.339	.297	.149	.298	.085	.136	.127	.136	.037
-.329	.137	.185	.259	.400	.099	.397	.010	.214	.061	.221	-.016
-.172	.104	.398	.141	.604	.010	.501	-.053	.292	.022	.295	-.071
-.030	.079	.737	.210	.785	.170	.603	.035	.403	.014	.396	-.143
.128	.055			.967	.098	.784	.065	.489	-.052	.497	-.204
.418	.136			1.000	-.480	.868	.230	.594	-.181	.597	-.192
.564	.162					.923	.289	.700	.018	.702	-.074
.710	.232					.972	.121	.786	.121	.786	.004
.976	.355							.858	.200	.864	.058
1.072	.290							.919	.208	.912	.049
1.110	.204							.967	.015	.985	-.322
CN=	1.0230		1.1712		.9848		.9854		.9811		.8046
CM=	.0202		-.1620		-.1441		-.1743		-.1921		-.1259

[redacted]

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Concluded.

$\alpha = 11.62^{\circ}$; $C_L = 1.117$

	STA	.123	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.425	-.021	-1.203	.023	-1.133	.025	-1.155	.022	-1.065	.018	-1.032
	-.567	-.569	.035	-1.200	.068	-1.103	.079	-1.146	.075	-1.034	.077	-1.134
	-.452	-.525	.105	-1.202	.209	-1.006	.133	-1.115	.129	-1.054	.129	-1.118
	-.311	-.531	.178	-1.146	.294	-.926	.214	-1.044	.201	-1.017	.209	-1.082
	-.023	-.548	.286	-1.104	.404	-.845	.295	-.916	.294	-.871	.293	-1.057
	.133	-.444	.396	-1.082	.497	-.801	.407	-.613	.397	-.856	.494	-.992
	.272	-.411	.514	-1.084	.599	-.755	.502	-.388	.495	-.823	.590	-.850
	.416	-.389	.618	-1.110	.700	-.715	.601	-.404	.594	-.808	.693	-.729
	.565	-.321	.733	-1.090	.864	-.666	.698	-.448	.693	-.752	.777	-.667
	.713	-.280	.835	-.938	.926	-.622	.863	-.461	.784	-.673	.861	-.613
	.854	-.333	.919	-.563			.923	-.463	.856	-.613	.918	-.596
	.980	-.361	.987	-.165			.977	-.453	.926	-.600	.972	-.588
	1.074	-.430							.977	-.601		
LOWER SURFACE												
	-.660	.082	-.022	.523	.024	.623	.074	.373	.019	.499	.020	.478
	-.616	.199	.038	.482	.075	.443	.130	.286	.066	.300	.076	.235
	-.462	.211	.101	.407	.297	.211	.298	.131	.136	.194	.136	.126
	-.329	.187	.185	.338	.400	.147	.397	.059	.214	.121	.221	.039
	-.172	.160	.398	.215	.604	.038	.501	.000	.292	.084	.295	-.024
	-.030	.138	.737	.239	.785	.185	.603	.061	.403	.044	.396	-.095
	.128	.117			.967	.088	.784	.081	.489	-.029	.497	-.178
	.418	.187			1.000	-.567	.868	.235	.594	-.158	.597	-.173
	.564	.211					.923	.297	.700	.032	.702	-.074
	.710	.276					.972	.113	.786	.124	.786	-.003
	.976	.396							.858	.209	.864	.045
	1.072	.320							.919	.210	.912	.045
	1.110	.220							.967	.013	.985	-.346
CN=	1.1707		1.3813		1.0319		.8212		.9195		.8718	
CM=	.0362		-.2292		-.1775		-.1184		-.1710		-.1344	

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TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(i) M = 1.00.

$\alpha = -1.06^{\circ}$; $C_L = -0.161$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.042	-.021	-.025	.023	-.087	.025	-.013	.022	.183	.018	.318
-.567	-.034	.035	-.223	.068	-.242	.079	-.134	.075	-.041	.077	.057
-.452	-.134	.105	-.249	.134	-.204	.133	-.136	.129	-.081	.129	.006
-.311	-.247	.178	-.321	.209	-.177	.214	-.132	.201	-.083	.209	-.033
-.023	-.346	.286	-.247	.294	-.177	.295	-.134	.294	-.072	.293	-.049
.133	-.262	.396	-.171	.404	-.157	.407	-.147	.397	-.039	.494	-.167
.272	-.221	.514	-.142	.497	-.179	.502	-.186	.495	-.100	.590	-.226
.416	-.172	.618	-.161	.599	-.234	.601	-.228	.594	-.136	.693	-.314
.565	-.042	.733	-.166	.700	-.275	.698	-.302	.693	-.217	.777	-.380
.713	-.007	.835	-.227	.864	-.203	.863	-.224	.784	-.309	.861	-.013
.854	-.058	.919	-.233	.926	-.025	.923	-.111	.856	-.301	.918	-.014
.980	-.112	.987	.003	.975	.039	.977	.043	.926	-.134	.972	-.039
1.074	-.161							.977	.019		
1.122	-.120										
				LOWER SURFACE							
-.660	.046	-.022	-.052	.024	-.345	.025	-.530	.019	-.629	.020	-.703
-.616	-.013	.038	-.326	.075	-.496	.130	-.755	.066	-.814	.076	-.665
-.572	-.066	.101	-.389	.297	-.539	.298	-.702	.136	-.843	.136	-.649
-.462	-.097	.185	-.423	.400	-.548	.397	-.683	.214	-.840	.221	-.538
-.329	-.117	.398	-.428	.604	-.213	.501	-.540	.292	-.761	.295	-.469
-.172	-.165	.737	-.068	.785	.079	.603	-.037	.403	-.740	.396	-.380
-.030	-.238			.967	.144	.703	.086	.489	-.492	.497	-.317
.128	-.355			1.000	.046	.784	.247	.594	-.194	.597	-.263
.418	-.290					.868	.306	.700	.152	.702	-.217
.564	-.297					.923	.238	.786	.232	.786	-.150
.710	-.200					.972	.189	.858	.313	.864	-.079
.976	.084							.919	.334	.912	-.046
1.072	.117							.967	.201		
1.110	.085										
CN=	-.0085		-.0868		-.0878		-.1297		-.2173		-.2399
CM=	.0052		-.0218		-.0434		-.0831		-.0903		-.0246

$\alpha = -0.07^{\circ}$; $C_L = -0.367$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.025	-.021	-.106	.023	-.247	.025	-.112	.022	.070	.018	.200
-.567	-.067	.035	-.388	.068	-.380	.079	-.235	.075	-.151	.077	-.007
-.452	-.166	.105	-.301	.134	-.344	.133	-.231	.129	-.166	.129	-.012
-.311	-.267	.178	-.386	.209	-.361	.214	-.198	.201	-.182	.209	-.033
-.023	-.368	.286	-.349	.294	-.278	.295	-.192	.294	-.154	.293	-.040
.133	-.289	.396	-.278	.404	-.184	.407	-.211	.397	-.161	.494	-.153
.272	-.247	.514	-.197	.497	-.193	.502	-.242	.495	-.220	.590	-.199
.416	-.212	.618	-.207	.599	-.263	.601	-.275	.594	-.042	.693	-.285
.565	-.093	.733	-.211	.700	-.302	.698	-.356	.693	-.111	.777	-.382
.713	-.047	.835	-.263	.864	-.327	.863	-.079	.784	-.219	.861	-.079
.854	-.089	.919	-.251	.926	-.063	.923	.001	.856	-.334	.918	.017
.980	-.143	.987	-.029	.975	-.010	.977	.069	.926	-.184	.972	.102
1.074	-.208							.977	.009		
1.122	-.158										
				LOWER SURFACE							
-.660	.053	-.022	.028	.024	-.218	.025	-.410	.019	-.497	.020	-.571
-.616	.006	.038	-.258	.075	-.417	.130	-.633	.066	-.742	.076	-.654
-.572	-.030	.101	-.321	.297	-.473	.298	-.579	.136	-.761	.136	-.577
-.462	-.070	.185	-.341	.400	-.528	.397	-.573	.214	-.728	.221	-.479
-.329	-.088	.398	-.396	.604	-.164	.501	-.466	.292	-.694	.295	-.361
-.172	-.148	.737	-.023	.785	.072	.603	.040	.403	-.382	.396	-.284
-.030	-.228			.967	.160	.703	.115	.489	-.045	.497	-.170
.128	-.333			1.000	-.007	.784	.165	.594	.011	.597	-.091
.418	-.265					.868	.255	.700	.137	.702	-.010
.564	-.269					.923	.231	.786	.235	.786	.061
.710	-.148					.972	.203	.858	.268	.864	.138
.976	.119							.919	.287	.912	.165
1.072	.129							.967	.214		
1.110	.088										
CN=	.0968		.0407		.0318		-.0404		-.0563		-.1005
CM=	-.0017		-.0370		-.0561		-.0710		-.1011		-.0674

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(i) M = 1.00. Continued.

$\alpha = 0.95^{\circ}$; $C_L = 0.090$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE											
-.660	.015	-.021	-.267	.023	-.364	.025	-.296	.022	-.113	.018	.111
-.567	-.094	.035	-.535	.068	-.502	.079	-.423	.075	-.250	.077	-.172
-.452	-.195	.105	-.413	.134	-.427	.133	-.438	.129	-.245	.129	-.185
-.311	-.288	.178	-.438	.209	-.438	.214	-.390	.201	-.250	.209	-.182
-.023	-.388	.286	-.382	.294	-.438	.295	-.326	.294	-.212	.293	-.190
.133	-.313	.396	-.371	.404	-.259	.407	-.202	.397	-.213	.494	-.070
.272	-.274	.514	-.248	.497	-.222	.502	-.262	.495	-.264	.590	-.161
.416	-.235	.618	-.245	.599	-.277	.601	-.301	.594	-.309	.693	-.246
.565	-.136	.733	-.247	.700	-.331	.698	-.372	.693	-.357	.777	-.323
.713	-.074	.835	-.281	.864	-.345	.863	-.133	.784	-.067	.861	-.060
.854	-.118	.919	-.254	.926	-.082	.923	-.019	.856	-.148	.918	-.010
.980	-.162	.987	-.038	.975	-.052	.977	.029	.926	-.117	.972	.093
1.074	-.231							.977	.033		
1.122	-.182										
LOWER SURFACE											
-.660	.066	-.022	.084	.024	-.114	.025	-.334	.019	-.381	.020	-.311
-.616	.022	.038	-.184	.075	-.341	.130	-.535	.066	-.616	.076	-.501
-.572	-.019	.101	-.280	.297	-.419	.298	-.501	.136	-.666	.136	-.415
-.462	-.054	.185	-.265	.400	-.441	.397	-.523	.214	-.587	.221	-.330
-.329	-.070	.398	-.356	.604	-.108	.501	-.203	.292	-.391	.295	-.215
-.172	-.134	.737	.009	.785	.121	.603	.061	.403	.063	.396	-.229
-.030	-.215			.967	.213	.703	.139	.489	-.006	.497	-.033
.128	-.310			1.000	-.036	.784	.180	.594	-.054	.597	.029
.418	-.247					.868	.220	.700	.113	.702	.143
.564	-.241					.923	.238	.786	.224	.786	.172
.710	-.108					.972	.193	.858	.283	.864	.208
.976	.137							.919	.302	.912	.216
1.072	.143							.967	.208		
1.110	.097										
CN=	.1790		.1509		.1561		.1014		.1006		.0569
CM=	-.0054		-.0447		-.0737		-.0795		-.1028		-.0768

$\alpha = 1.97^{\circ}$; $C_L = 0.219$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE											
-.660	.007	-.021	-.426	.023	-.461	.025	-.430	.022	-.237	.018	-.164
-.567	-.132	.035	-.670	.068	-.642	.079	-.550	.075	-.497	.077	-.404
-.452	-.211	.105	-.499	.134	-.590	.133	-.555	.129	-.463	.129	-.429
-.311	-.307	.178	-.517	.209	-.515	.214	-.559	.201	-.509	.209	-.380
-.023	-.402	.286	-.426	.294	-.486	.295	-.471	.294	-.409	.293	-.288
.133	-.340	.396	-.440	.404	-.450	.407	-.413	.397	-.321	.494	-.281
.272	-.290	.514	-.311	.497	-.331	.502	-.276	.495	-.225	.590	-.279
.416	-.262	.618	-.287	.599	-.288	.601	-.303	.594	-.283	.693	-.078
.565	-.173	.733	-.278	.700	-.388	.698	-.381	.693	-.379	.777	-.110
.713	-.099	.835	-.312	.864	-.300	.863	-.161	.784	-.382	.861	-.129
.854	-.137	.919	-.223	.926	-.100	.923	-.046	.856	-.153	.918	-.062
.980	-.182	.987	-.051	.975	-.083	.977	-.020	.926	-.067	.972	.085
1.074	-.252							.977	.043		
1.122	-.199										
LOWER SURFACE											
-.660	.078	-.022	.151	.024	-.013	.025	-.238	.019	-.245	.020	-.034
-.616	.044	.038	-.130	.075	-.252	.130	-.392	.066	-.466	.076	-.319
-.572	.005	.101	-.177	.297	-.368	.298	-.400	.136	-.383	.136	-.351
-.462	-.024	.185	-.228	.400	-.389	.397	-.198	.214	-.351	.221	-.292
-.329	-.054	.398	-.315	.604	-.104	.501	-.186	.292	-.045	.295	-.294
-.172	-.116	.737	.050	.785	.149	.603	.046	.403	-.162	.396	-.163
-.030	-.159			.967	.215	.703	.099	.489	-.119	.497	-.048
.128	-.276			1.000	-.054	.784	.166	.594	-.194	.597	.055
.418	-.222					.868	.224	.700	.086	.702	.174
.564	-.204					.923	.250	.786	.199	.786	.244
.710	-.070					.972	.193	.858	.260	.864	.288
.976	.169							.919	.290	.912	.291
1.072	.166							.967	.214		
1.110	.106										
CN=	.2676		.2599		.2663		.2526		.2356		.2014
CM=	-.0085		-.0538		-.0796		-.0848		-.0904		-.0788

TABLE IV — Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(i) M = 1.00. Continued.

$\alpha = 2.47^{\circ}$; $C_L = 0.287$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.011	-.021	-.512	.023	-.516	.025	-.483	.022	-.305	.018	-.250
-.567	-.147	.035	-.700	.068	-.682	.079	-.591	.075	-.538	.077	-.500
-.452	-.227	.105	-.638	.134	-.660	.133	-.611	.129	-.536	.129	-.542
-.311	-.319	.178	-.531	.209	-.606	.214	-.611	.201	-.550	.209	-.499
-.023	-.418	.286	-.446	.294	-.565	.295	-.547	.294	-.495	.293	-.464
.133	-.347	.396	-.459	.404	-.461	.407	-.495	.397	-.481	.494	-.382
.272	-.295	.514	-.353	.497	-.459	.502	-.442	.495	-.393	.590	-.302
.416	-.278	.618	-.307	.599	-.296	.601	-.305	.594	-.277	.693	-.073
.565	-.188	.733	-.295	.700	-.419	.698	-.379	.693	-.364	.777	-.004
.713	-.124	.835	-.322	.864	-.283	.863	-.170	.784	-.449	.861	-.044
.854	-.158	.919	-.211	.926	-.106	.923	-.065	.856	-.172	.918	-.037
.980	-.197	.987	-.054	.975	-.091	.977	-.028	.926	-.060	.972	.089
1.074	-.267							.977	.018		
1.122	-.207										
				LOWER SURFACE							
-.660	.078	-.022	.174	.024	.038	.025	-.145	.019	-.072	.020	.001
-.616	.050	.038	-.078	.075	-.208	.130	-.306	.066	-.299	.076	-.308
-.572	.018	.101	-.144	.297	-.311	.298	-.135	.136	-.242	.136	-.330
-.462	-.020	.185	-.207	.400	-.322	.397	-.211	.214	-.242	.221	-.247
-.329	-.051	.398	-.284	.604	-.115	.501	-.172	.292	-.172	.295	-.264
-.172	-.110	.737	.062	.785	.157	.603	.022	.403	-.212	.396	-.193
-.030	-.200			.967	.206	.703	.076	.489	-.084	.497	-.184
.128	-.270			1.000	-.071	.784	.146	.594	-.192	.597	-.026
.418	-.216					.868	.217	.700	.079	.702	.155
.564	-.183					.923	.248	.786	.201	.786	.239
.710	-.044					.972	.188	.858	.266	.864	.276
.976	.183							.919	.295	.912	.285
1.072	.173							.967	.195		
1.110	.114										
CN=	.3114		.3150		.3340		.3472		.3175		.2345
CM=	-.0131		-.0564		-.0824		-.0862		-.0929		-.0621

$\alpha = 2.96^{\circ}$; $C_L = 0.353$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.016	-.021	-.586	.023	-.558	.025	-.512	.022	-.380	.018	-.314
-.567	-.160	.035	-.722	.068	-.720	.079	-.619	.075	-.618	.077	-.546
-.452	-.234	.105	-.654	.134	-.702	.133	-.641	.129	-.590	.129	-.591
-.311	-.325	.178	-.551	.209	-.684	.214	-.643	.201	-.603	.209	-.561
-.023	-.422	.286	-.459	.294	-.629	.295	-.590	.294	-.541	.293	-.528
.133	-.354	.396	-.480	.404	-.501	.407	-.554	.397	-.539	.494	-.515
.272	-.306	.514	-.371	.497	-.471	.502	-.566	.495	-.534	.590	-.509
.416	-.285	.618	-.322	.599	-.315	.601	-.364	.594	-.551	.693	-.269
.565	-.194	.733	-.307	.700	-.422	.698	-.379	.693	-.419	.777	-.007
.713	-.123	.835	-.324	.864	-.280	.863	-.193	.784	-.410	.861	.026
.854	-.165	.919	-.183	.926	-.106	.923	-.084	.856	-.165	.918	.027
.980	-.208	.987	-.053	.975	-.089	.977	-.051	.926	-.056	.972	.104
1.074	-.270							.977	-.012		
1.122	-.197										
				LOWER SURFACE							
-.660	.087	-.022	.200	.024	.069	.025	-.057	.019	-.053	.020	.040
-.616	.061	.038	-.043	.075	-.168	.130	-.156	.066	-.255	.076	-.248
-.572	.027	.101	-.123	.297	-.287	.298	-.167	.136	-.272	.136	-.221
-.462	-.010	.185	-.192	.400	-.245	.397	-.235	.214	-.221	.221	-.240
-.329	-.041	.398	-.246	.604	-.130	.501	-.194	.292	-.190	.295	-.272
-.172	-.108	.737	.081	.785	.171	.603	.021	.403	-.189	.396	-.194
-.030	-.195			.967	.196	.703	.082	.489	-.074	.497	-.193
.128	-.249			1.000	-.076	.784	.149	.594	-.203	.597	-.046
.418	-.202					.868	.212	.700	.065	.702	.116
.564	-.161					.923	.246	.786	.189	.786	.195
.710	-.028					.972	.182	.858	.261	.864	.236
.976	.197							.919	.285	.912	.249
1.072	.190							.967	.174		
1.110	.128										
CN=	.3465		.3565		.3816		.4057		.3865		.3037
CM=	-.0142		-.0601		-.0830		-.0892		-.1028		-.0642

[redacted]

TABLE IV – Continued

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(i) M = 1.00. Continued.

$\alpha = 3.47^{\circ}$; $C_L = 0.418$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.031	-.021	-.677	.023	-.613	.025	-.580	.022	-.446	.018	-.368
-.567	-.188	.035	-.768	.C68	-.761	.079	-.680	.075	-.664	.077	-.611
-.452	-.249	.105	-.761	.134	-.752	.133	-.690	.129	-.654	.129	-.640
-.311	-.340	.178	-.582	.209	-.742	.214	-.701	.201	-.664	.209	-.627
-.023	-.437	.286	-.482	.294	-.695	.295	-.641	.294	-.623	.293	-.592
.133	-.370	.396	-.497	.404	-.577	.407	-.634	.397	-.604	.494	-.581
.272	-.321	.514	-.398	.497	-.479	.502	-.632	.495	-.602	.590	-.584
.416	-.295	.618	-.344	.599	-.38C	.601	-.599	.594	-.601	.693	-.206
.565	-.217	.733	-.323	.700	-.434	.698	-.433	.693	-.633	.777	-.083
.713	-.145	.835	-.330	.864	-.28C	.863	-.189	.784	-.256	.861	-.034
.854	-.173	.919	-.163	.926	-.118	.923	-.109	.856	-.176	.918	.010
.980	-.222	.987	-.063	.975	-.101	.977	-.092	.926	-.159	.972	.018
1.074	-.281							.977	-.153		
1.122	-.2C6										
					LOWER SURFACE						
-.660	.C93	-.022	.236	.024	.142	.025	.062	.019	.027	.020	.083
-.616	.073	.038	.002	.C75	-.105	.130	-.163	.066	-.215	.076	-.173
-.572	.040	.101	-.089	.297	-.16C	.298	-.145	.136	-.208	.136	-.201
-.462	.0C2	.185	-.146	.400	-.138	.397	-.212	.214	-.200	.221	-.215
-.329	-.03C	.398	-.200	.604	-.122	.501	-.185	.292	-.117	.295	-.259
-.172	-.093	.737	.109	.785	.174	.603	.024	.403	-.201	.396	-.215
-.030	-.178			.967	.195	.703	.064	.489	-.09C	.497	-.249
.128	-.240			1.000	-.087	.784	.134	.594	-.198	.597	-.109
.418	-.186					.868	.221	.700	.054	.702	.045
.564	-.129					.923	.267	.786	.168	.786	.139
.710	.C11					.972	.179	.858	.245	.864	.182
.976	.227							.919	.259	.912	.192
1.072	.210							.967	.116		
1.110	.145										
CN=	.4104		.4268		.47C5		.4867		.4571		.3311
CM=	-.0186		-.0655		-.C888		-.1025		-.1103		-.0579

$\alpha = 3.95^{\circ}$; $C_L = 0.476$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.056	-.021	-.702	.023	-.66C	.025	-.624	.022	-.497	.018	-.448
-.567	-.214	.035	-.857	.068	-.804	.079	-.732	.075	-.701	.077	-.649
-.452	-.266	.1C5	-.814	.134	-.790	.133	-.735	.129	-.689	.129	-.686
-.311	-.347	.178	-.649	.209	-.795	.214	-.739	.201	-.715	.209	-.675
-.023	-.448	.286	-.495	.294	-.743	.295	-.702	.294	-.660	.293	-.635
.133	-.378	.396	-.510	.404	-.716	.407	-.686	.397	-.645	.494	-.635
.272	-.326	.514	-.427	.497	-.515	.502	-.669	.495	-.630	.590	-.630
.416	-.315	.618	-.368	.599	-.431	.601	-.675	.594	-.647	.693	-.229
.565	-.232	.733	-.337	.700	-.454	.698	-.669	.693	-.576	.777	-.132
.713	-.154	.835	-.328	.864	-.264	.863	-.196	.784	-.240	.861	-.106
.854	-.195	.919	-.162	.926	-.130	.923	-.178	.856	-.212	.918	-.076
.98C	-.234	.987	-.071	.975	-.121	.977	-.142	.926	-.209	.972	-.077
1.074	-.293							.977	-.210		
1.122	-.211										
					LOWER SURFACE						
-.660	.09C	-.022	.265	.024	.216	.025	.117	.019	.08C	.020	.131
-.616	.082	.038	.027	.075	-.009	.130	-.069	.066	-.152	.076	-.108
-.572	.054	.101	-.069	.297	-.108	.298	-.163	.136	-.168	.136	-.189
-.462	.0C7	.185	-.138	.400	-.122	.397	-.213	.214	-.110	.221	-.204
-.329	-.C24	.398	-.088	.604	-.097	.501	-.166	.292	-.133	.295	-.241
-.172	-.091	.737	.117	.785	.163	.603	.033	.403	-.200	.396	-.258
-.030	-.176			.967	.183	.703	.058	.489	-.080	.497	-.254
.128	-.230			1.000	-.103	.784	.120	.594	-.197	.597	-.159
.418	-.163					.868	.219	.700	.029	.702	.012
.564	-.1C2					.923	.268	.786	.148	.786	.1C6
.710	.055					.972	.161	.858	.222	.864	.148
.976	.234							.919	.239	.912	.164
1.072	.210							.967	.084		
1.110	.141										
CN=	.4593		.4947		.54C8		.5729		.4919		.3722
CM=	-.0217		-.C713		-.0912		-.1209		-.1079		-.C631

[redacted]

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TABLE IV – Concluded

PRESSURE DISTRIBUTIONS OVER WING WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(i) M = 1.00. Concluded.

$\alpha = 4.96^{\circ}$; $C_L = 0.586$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.107	-.021	-.799	.023	-.762	.025	-.727	.022	-.618	.018	-.552
-.567	-.239	.035	-.927	.068	-.888	.079	-.800	.075	-.783	.077	-.735
-.452	-.299	.105	-.898	.134	-.865	.133	-.824	.129	-.778	.129	-.763
-.311	-.368	.178	-.876	.209	-.857	.214	-.817	.201	-.779	.209	-.752
-.023	-.473	.286	-.529	.294	-.821	.295	-.775	.294	-.746	.293	-.716
.133	-.397	.396	-.535	.404	-.808	.407	-.768	.397	-.740	.494	-.698
.272	-.346	.514	-.459	.497	-.787	.502	-.758	.495	-.725	.590	-.688
.416	-.330	.618	-.410	.599	-.520	.601	-.754	.594	-.728	.693	-.265
.565	-.240	.733	-.356	.700	-.515	.698	-.764	.693	-.310	.777	-.224
.713	-.168	.835	-.325	.864	-.221	.863	-.341	.784	-.283	.861	-.188
.854	-.203	.919	-.156	.926	-.162	.923	-.334	.856	-.277	.918	-.178
.980	-.241	.987	-.081	.975	-.146	.977	-.328	.926	-.274	.972	-.191
1.074	-.306							.977	-.275		
1.122	-.232										
				LOWER SURFACE							
-.660	.097	-.022	.338	.024	.300	.025	.245	.019	.190	.020	.232
-.616	.093	.038	.132	.075	.082	.130	-.006	.066	-.030	.076	-.066
-.572	.077	.101	.061	.297	-.062	.298	-.129	.136	-.060	.136	-.138
-.462	.034	.185	-.002	.400	-.069	.397	-.037	.214	-.079	.221	-.172
-.329	-.005	.398	-.051	.604	-.031	.501	-.146	.292	-.115	.295	-.215
-.172	-.066	.737	.134	.785	.170	.603	.031	.403	-.177	.396	-.269
-.030	-.141			.967	.181	.703	.047	.489	-.050	.497	-.242
.128	-.201			1.000	-.115	.784	.106	.594	-.198	.597	-.184
.418	-.065					.868	.221	.700	-.006	.702	-.023
.564	-.011					.923	.273	.786	.126	.786	.057
.710	.100					.972	.140	.858	.203	.864	.109
.976	.264							.919	.220	.912	.111
1.072	.228							.967	.061		
1.110	.152										
CN=	.5688		.6111		.6651		.7134		.5582		.4466
CM=	-.0219		-.0705		-.1064		-.1516		-.1029		-.0707

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TABLE V

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25.

$\alpha = -5.08^o$; $C_L = -0.411$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.036	-.021	.325	.023	.426	.025	.428	.022	.504	.018	.514
-.567	.019	.035	.189	.068	.272	.079	.281	.075	.310	.077	.279
-.452	-.050	.105	.129	.134	.153	.133	.197	.129	.237	.129	.233
-.311	-.082	.178	.057	.209	.095	.214	.129	.201	.176	.209	.162
-.023	-.066	.286	.009	.294	.059	.295	.098	.294	.127	.293	.096
.133	-.026	.396	-.008	.404	.027	.407	.042	.397	.081	.494	.007
.272	.004	.514	-.020	.497	.005	.502	.011	.495	.025	.590	-.025
.416	.016	.618	-.036	.599	-.042	.601	-.025	.594	-.017	.693	-.074
.565	.024	.733	-.048	.700	-.068	.698	-.067	.693	-.068	.777	-.111
.713	.015	.835	-.066	.864	-.128	.863	-.149	.764	-.109	.861	-.123
.854	-.009	.919	-.069	.926	-.106	.923	-.133	.856	-.137	.918	-.138
.960	-.017	.987	-.046	.975	-.050	.977	-.057	.926	-.134	.972	-.068
1.074	-.029							.977	-.047		
1.122	-.020										
					LOWER SURFACE						
-.660	-.098	-.022	-1.101	.024	-1.527	.025	-1.956	.019	-2.773	.020	-2.535
-.616	-.197	.038	-.744	.075	-.929	.130	-.715	.066	-1.184	.076	-1.048
-.572	-.236	.101	-.627	.297	-.425	.298	-.407	.136	-.764	.136	-.748
-.462	-.256	.185	-.501	.400	-.328	.397	-.326	.214	-.538	.221	-.482
-.329	-.213	.398	-.315	.604	-.196	.501	-.246	.292	-.429	.295	-.366
-.172	-.197	.737	.017	.785	.057	.603	-.104	.403	-.328	.396	-.251
-.030	-.259			.967	.159	.703	.002	.489	-.281	.497	-.170
.128	-.309			1.000	.043	.784	.049	.594	-.191	.597	-.076
.418	-.195					.868	.132	.700	-.061	.702	-.005
.564	-.127					.923	.128	.786	.033	.786	.032
.710	-.015					.972	.127	.858	.083	.864	.035
.976	.165							.919	.124	.912	.040
1.072	.169							.967	.114		
1.110	.163										
CN=	-.2107		-.3421		-.3547		-.3812		-.4464		-.3927
CM=	-.1067		-.0499		-.0605		-.0752		-.0692		-.0711

$\alpha = -4.12^o$; $C_L = -0.329$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.035	-.021	.291	.023	.356	.025	.370	.022	.470	.018	.496
-.567	.004	.035	.126	.068	.194	.079	.219	.075	.257	.077	.215
-.452	-.068	.105	.057	.134	.104	.133	.153	.129	.191	.129	.186
-.311	-.103	.178	-.008	.209	.056	.214	.085	.201	.121	.209	.114
-.023	-.093	.286	-.016	.294	.021	.295	.050	.294	.085	.293	.067
.133	-.040	.396	-.036	.404	-.013	.407	.014	.397	.045	.494	-.009
.272	-.024	.514	-.062	.497	-.041	.502	-.021	.495	.000	.590	-.052
.416	-.005	.618	-.041	.599	-.078	.601	-.053	.594	-.033	.693	-.082
.565	.017	.733	-.069	.700	-.101	.698	-.092	.693	-.079	.777	-.121
.713	.005	.835	-.090	.864	-.142	.863	-.153	.784	-.121	.861	-.139
.854	-.017	.919	-.070	.926	-.123	.923	-.142	.856	-.162	.918	-.129
.980	-.026	.987	-.052	.975	-.057	.977	-.059	.926	-.142	.972	-.055
1.074	-.044							.977	-.051		
1.122	-.030										
					LOWER SURFACE						
-.660	-.065	-.022	-.861	.024	-1.291	.025	-1.695	.019	-1.997	.020	-2.231
-.616	-.182	.038	-.667	.075	-.800	.130	-.635	.066	-1.048	.076	-.927
-.572	-.207	.101	-.531	.297	-.385	.298	-.380	.136	-.663	.136	-.679
-.462	-.235	.185	-.455	.400	-.313	.397	-.298	.214	-.488	.221	-.453
-.329	-.201	.398	-.283	.604	-.194	.501	-.223	.292	-.360	.295	-.334
-.172	-.195	.737	.015	.785	.052	.603	-.100	.403	-.295	.396	-.254
-.030	-.242			.967	.155	.703	.007	.489	-.249	.497	-.180
.128	-.283			1.000	.033	.784	.052	.594	-.174	.597	-.090
.418	-.191					.868	.139	.700	-.027	.702	.007
.564	-.124					.923	.152	.786	.058	.786	.048
.710	-.014					.972	.116	.858	.102	.864	.052
.976	.156							.919	.136	.912	.075
1.072	.148							.967	.120		
1.110	.114										
CN=	-.1678		-.2698		-.2803		-.3086		-.3439		-.3344
CM=	-.0918		-.0474		-.0601		-.0747		-.0681		-.0706

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(a) M = 0.25. Continued.

$\alpha = -3.10^{\circ}$; $C_L = -0.234$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.035	-.021	.238	.023	.256	.025	.289	.022	.383	.018	.451
-.567	-.004	.035	.044	.068	.131	.079	.155	.075	.186	.077	.120
-.452	-.081	.105	-.006	.134	.013	.133	.093	.129	.116	.129	.124
-.311	-.125	.178	-.054	.209	-.001	.214	.037	.201	.064	.209	.065
-.023	-.106	.286	-.062	.294	-.032	.295	.014	.294	.041	.293	.034
.133	-.066	.396	-.082	.404	-.046	.407	-.020	.397	.000	.494	-.037
.272	-.038	.514	-.068	.497	-.058	.502	-.052	.495	-.032	.590	-.060
.416	-.024	.618	-.078	.599	-.080	.601	-.080	.594	-.070	.693	-.088
.565	-.020	.733	-.080	.700	-.109	.698	-.106	.693	-.110	.777	-.123
.713	-.006	.835	-.081	.864	-.145	.863	-.173	.784	-.139	.861	-.130
.854	-.016	.919	-.084	.926	-.126	.923	-.150	.856	-.164	.918	-.133
.980	-.041	.987	-.045	.975	-.061	.977	-.055	.926	-.153	.972	-.051
1.074	-.033							.977	-.045		
1.122	-.013										
					LOWER SURFACE						
-.660	-.037	-.022	-.548	.024	-.918	.025	-1.270	.019	-1.449	.020	-1.692
-.616	-.146	.038	-.560	.075	-.677	.130	-.545	.066	-.794	.076	-.821
-.572	-.160	.101	-.488	.297	-.326	.298	-.310	.136	-.524	.136	-.566
-.462	-.209	.165	-.355	.400	-.265	.397	-.268	.214	-.412	.221	-.381
-.329	-.163	.398	-.246	.604	-.169	.501	-.201	.292	-.326	.295	-.284
-.172	-.160	.737	.036	.785	.075	.603	-.096	.403	-.258	.396	-.227
-.030	-.213			.967	.162	.703	.027	.489	-.224	.497	-.162
.128	-.259			1.000	.038	.784	.079	.594	-.152	.597	-.080
.418	-.160					.868	.139	.700	-.013	.702	.040
.564	-.095					.923	.162	.786	.088	.786	.097
.710	.011					.972	.132	.858	.135	.864	.117
.976	.176							.919	.160	.912	.122
1.072	.151							.967	.134		
1.110	.118										
CN=	-.0911		-.1721		-.1875		-.2108		-.2319		-.2366
CM=	-.0818		-.0465		-.0588		-.0732		-.0682		-.0729

$\alpha = -2.08^{\circ}$; $C_L = -0.142$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.033	-.021	.157	.023	.144	.025	.168	.022	.311	.018	.388
-.567	-.036	.035	-.008	.068	.037	.079	.058	.075	.109	.077	.061
-.452	-.122	.105	-.068	.134	-.045	.133	.010	.129	.050	.129	.080
-.311	-.162	.178	-.104	.209	-.076	.214	-.016	.201	.023	.209	.031
-.023	-.147	.286	-.113	.294	-.078	.295	-.040	.294	.000	.293	-.012
.133	-.082	.396	-.111	.404	-.092	.407	-.063	.397	-.025	.494	-.059
.272	-.068	.514	-.105	.497	-.101	.502	-.079	.495	-.068	.590	-.092
.416	-.045	.618	-.104	.599	-.111	.601	-.108	.594	-.095	.693	-.125
.565	-.021	.733	-.101	.700	-.139	.698	-.153	.693	-.137	.777	-.149
.713	-.022	.835	-.107	.864	-.163	.863	-.185	.784	-.166	.861	-.146
.854	-.033	.919	-.097	.926	-.138	.923	-.160	.856	-.182	.918	-.142
.980	-.045	.987	-.052	.975	-.065	.977	-.060	.926	-.160	.972	-.057
1.074	-.048							.977	-.055		
1.122	-.024										
					LOWER SURFACE						
-.660	-.011	-.022	-.396	.024	-.671	.025	-.968	.019	-1.147	.020	-1.072
-.616	-.101	.038	-.423	.075	-.529	.130	-.415	.066	-.710	.076	-.654
-.572	-.130	.101	-.370	.297	-.272	.298	-.268	.136	-.463	.136	-.468
-.462	-.140	.185	-.321	.400	-.211	.397	-.218	.214	-.332	.221	-.319
-.329	-.140	.398	-.209	.604	-.145	.501	-.176	.292	-.270	.295	-.246
-.172	-.138	.737	.051	.785	.101	.603	-.077	.403	-.215	.396	-.194
-.030	-.176			.967	.160	.703	.026	.489	-.194	.497	-.146
.128	-.226			1.000	.034	.784	.102	.594	-.139	.597	-.071
.418	-.137					.868	.137	.700	.004	.702	.048
.564	-.076					.923	.176	.786	.101	.786	.131
.710	.024					.972	.134	.858	.158	.864	.163
.976	.173							.919	.184	.912	.153
1.072	.152							.967	.129		
1.110	.121										
CN=	-.0064		-.0864		-.0857		-.1076		-.1496		-.1337
CM=	-.0593		-.0493		-.0623		-.0733		-.0723		-.0750

[redacted]

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(a) M = 0.25. Continued.

$\alpha = -1.09^{\circ}$; $C_L = -0.050$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.029	-.021	.072	.023	-.014	.025	.072	.022	.209	.018	.289
-.567	-.074	.035	-.125	.068	-.065	.079	-.023	.075	.029	.077	-.060
-.452	-.158	.105	-.139	.134	-.118	.133	-.049	.129	-.002	.129	.012
-.311	-.185	.178	-.167	.209	-.105	.214	-.068	.201	-.028	.209	-.024
-.023	-.158	.286	-.153	.294	-.112	.295	-.079	.294	-.047	.293	-.044
.133	-.105	.396	-.145	.404	-.122	.407	-.083	.397	-.056	.494	-.076
.272	-.090	.514	-.126	.497	-.132	.502	-.106	.495	-.091	.590	-.103
.416	-.048	.618	-.132	.599	-.139	.601	-.126	.594	-.123	.693	-.135
.565	-.036	.733	-.114	.700	-.162	.698	-.156	.693	-.153	.777	-.157
.713	-.038	.835	-.122	.864	-.174	.863	-.199	.784	-.185	.861	-.149
.854	-.052	.919	-.101	.926	-.145	.923	-.158	.856	-.207	.918	-.142
.980	-.054	.987	-.057	.975	-.061	.977	-.056	.926	-.166	.972	-.056
1.074	-.050							.977	-.053		
1.122	-.021										
					LOWER SURFACE						
-.660	.015	-.022	-.161	.024	-.467	.025	-.644	.019	-.852	.020	-.738
-.616	-.057	.038	-.297	.075	-.418	.130	-.341	.066	-.527	.076	-.567
-.572	-.109	.101	-.314	.297	-.223	.298	-.225	.136	-.371	.136	-.438
-.462	-.122	.185	-.259	.400	-.175	.357	-.197	.214	-.250	.221	-.263
-.329	-.105	.398	-.181	.604	-.124	.501	-.151	.292	-.218	.295	-.206
-.172	-.108	.737	.067	.785	.118	.603	-.061	.403	-.192	.396	-.175
-.030	-.167			.967	.165	.703	.038	.489	-.160	.497	-.124
.128	-.197			1.000	.035	.784	.113	.594	-.121	.597	-.063
.418	-.112					.868	.161	.700	.015	.702	.057
.564	-.058					.923	.184	.786	.116	.786	.136
.710	.037					.972	.134	.858	.180	.864	.190
.976	.181							.919	.207	.912	.201
1.072	.181							.967	.136		
1.110	.123										
CN=	.0635		.0034		-.0023		-.0273		-.0610		-.0588
CM=	-.0474		-.0484		-.0634		-.0707		-.0740		-.0756

(a) M = 0.25. Continued.

$\alpha = -0.05^{\circ}$; $C_L = 0.043$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.008	-.021	-.078	.023	-.166	.025	-.091	.022	.039	.018	.187
-.567	-.110	.035	-.242	.068	-.193	.079	-.156	.075	-.070	.077	-.165
-.452	-.178	.105	-.218	.134	-.197	.133	-.129	.129	-.080	.129	-.044
-.311	-.222	.178	-.225	.209	-.183	.214	-.130	.201	-.111	.209	-.074
-.023	-.187	.286	-.219	.294	-.177	.295	-.133	.294	-.103	.293	-.091
.133	-.134	.396	-.179	.404	-.157	.407	-.133	.397	-.104	.494	-.103
.272	-.109	.514	-.151	.497	-.155	.502	-.152	.495	-.123	.590	-.122
.416	-.062	.618	-.154	.599	-.162	.601	-.152	.594	-.145	.693	-.152
.565	-.057	.733	-.131	.700	-.181	.698	-.180	.693	-.172	.777	-.184
.713	-.045	.835	-.127	.864	-.187	.863	-.204	.784	-.193	.861	-.166
.854	-.061	.919	-.112	.926	-.147	.923	-.165	.856	-.205	.918	-.152
.980	-.061	.987	-.053	.975	-.069	.977	-.057	.926	-.172	.972	-.064
1.074	-.054							.977	-.057		
1.122	-.030										
					LOWER SURFACE						
-.660	.038	-.022	-.035	.024	-.235	.025	-.342	.019	-.490	.020	-.383
-.616	-.010	.038	-.182	.075	-.300	.130	-.268	.066	-.407	.076	-.415
-.572	-.061	.101	-.229	.297	-.194	.298	-.174	.136	-.289	.136	-.341
-.462	-.087	.185	-.180	.400	-.158	.397	-.145	.214	-.225	.221	-.235
-.329	-.078	.398	-.137	.604	-.106	.501	-.128	.292	-.177	.295	-.164
-.172	-.081	.737	.087	.785	.140	.603	-.053	.403	-.147	.396	-.150
-.030	-.149			.967	.162	.703	.035	.489	-.131	.497	-.110
.128	-.176			1.000	.033	.784	.130	.594	-.095	.597	-.051
.418	-.083					.868	.191	.700	.038	.702	.064
.564	-.031					.923	.228	.786	.142	.786	.153
.710	.064					.972	.160	.858	.216	.864	.206
.976	.197							.919	.238	.912	.218
1.072	.168							.967	.147		
1.110	.125										
CN=	.1508		.1037		.0843		.0753		.0363		.0271
CM=	-.0328		-.0486		-.0628		-.0720		-.0762		-.0760

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(a) M = 0.25. Continued.

$\alpha = 0.98^{\circ}$; $C_L = 0.132$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.008	-.021	-.226	.023	-.335	.025	-.209	.022	-.178	.018	-.018
-.567	-.133	.035	-.338	.068	-.283	.079	-.239	.075	-.200	.077	-.270
-.452	-.222	.105	-.295	.134	-.286	.133	-.215	.129	-.165	.129	-.131
-.311	-.250	.178	-.271	.209	-.221	.214	-.196	.201	-.165	.209	-.126
-.023	-.225	.286	-.249	.294	-.222	.295	-.185	.294	-.160	.293	-.115
.133	-.154	.396	-.218	.404	-.198	.407	-.173	.397	-.139	.494	-.133
.272	-.124	.514	-.177	.497	-.190	.502	-.182	.495	-.161	.590	-.129
.416	-.094	.618	-.161	.599	-.185	.601	-.186	.594	-.178	.693	-.166
.565	-.065	.733	-.137	.700	-.194	.698	-.216	.693	-.196	.777	-.185
.713	-.057	.835	-.138	.864	-.198	.863	-.217	.784	-.212	.861	-.180
.854	-.070	.919	-.110	.926	-.157	.923	-.178	.856	-.221	.918	-.169
.980	-.066	.987	-.048	.975	-.069	.977	-.066	.926	-.182	.972	-.072
1.074	-.053							.977	-.064		
1.122	-.025										
					LOWER SURFACE						
-.660	.061	-.022	.118	.024	-.048	.025	-.128	.019	-.246	.020	-.227
-.616	.013	.038	-.116	.075	-.167	.130	-.197	.066	-.287	.076	-.294
-.572	-.026	.101	-.153	.297	-.150	.298	-.162	.136	-.201	.136	-.248
-.462	-.072	.185	-.140	.400	-.124	.397	-.114	.214	-.170	.221	-.198
-.329	-.068	.398	-.113	.604	-.095	.501	-.098	.292	-.153	.295	-.154
-.172	-.069	.737	.088	.785	.142	.603	-.031	.403	-.105	.396	-.102
-.030	-.117			.967	.165	.703	.033	.489	-.104	.497	-.085
.128	-.142			1.000	.021	.784	.131	.594	-.085	.597	-.043
.418	-.074					.868	.200	.700	.046	.702	.066
.564	-.023					.923	.236	.786	.151	.786	.158
.710	.074					.972	.161	.858	.224	.864	.209
.976	.200							.919	.244	.912	.219
1.072	.195							.967	.138		
1.110	.124										
CN=	.2127		.1710		.1661		.1573		.1256		.0992
CM=	-.0154		-.0437		-.0609		-.0737		-.0757		-.0738

$\alpha = 1.97^{\circ}$; $C_L = 0.216$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.042	-.021	-.389	.023	-.576	.025	-.425	.022	-.302	.018	-.188
-.567	-.178	.035	-.434	.068	-.382	.079	-.358	.075	-.296	.077	-.385
-.452	-.246	.105	-.355	.134	-.355	.133	-.295	.129	-.241	.129	-.188
-.311	-.281	.178	-.341	.209	-.308	.214	-.255	.201	-.220	.209	-.166
-.023	-.239	.286	-.285	.294	-.258	.295	-.220	.294	-.200	.293	-.161
.133	-.177	.396	-.259	.404	-.227	.407	-.210	.397	-.178	.494	-.154
.272	-.146	.514	-.200	.497	-.211	.502	-.208	.495	-.190	.590	-.161
.416	-.113	.618	-.187	.599	-.208	.601	-.220	.594	-.200	.693	-.184
.565	-.084	.733	-.166	.700	-.206	.698	-.231	.693	-.221	.777	-.205
.713	-.069	.835	-.153	.864	-.200	.863	-.232	.784	-.235	.861	-.192
.854	-.078	.919	-.102	.926	-.155	.923	-.177	.856	-.233	.918	-.172
.980	-.076	.987	-.049	.975	-.072	.977	-.074	.926	-.194	.972	-.072
1.074	-.052							.977	-.071		
1.122	-.036										
					LOWER SURFACE						
-.660	.070	-.022	.208	.024	.118	.025	.061	.019	-.016	.020	.000
-.616	.024	.038	-.009	.075	-.064	.130	-.110	.066	-.154	.076	-.160
-.572	-.007	.101	-.063	.297	-.101	.298	-.115	.136	-.139	.136	-.173
-.462	-.039	.185	-.076	.400	-.087	.397	-.082	.214	-.110	.221	-.141
-.329	-.040	.398	-.087	.604	-.074	.501	-.072	.292	-.093	.295	-.127
-.172	-.046	.737	.090	.785	.153	.603	-.026	.403	-.076	.396	-.088
-.030	-.089			.967	.149	.703	.056	.489	-.083	.497	-.083
.128	-.125			1.000	.016	.784	.126	.594	-.067	.597	-.039
.418	-.049					.868	.211	.700	.047	.702	.065
.564	-.002					.923	.248	.786	.145	.786	.152
.710	.085					.972	.155	.858	.220	.864	.209
.976	.206							.919	.234	.912	.218
1.072	.196							.967	.136		
1.110	.131										
CN=	.2821		.2521		.2473		.2438		.2032		.1700
CM=	-.0005		-.0409		-.0577		-.0721		-.0747		-.0701

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(a) M = 0.25. Continued.

$\alpha = 2.48^{\circ}$; $C_L = 0.260$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.066	-.021	-.501	.023	-.696	.025	-.493	.022	-.436	.018	-.299
-.567	-.181	.035	-.480	.068	-.460	.079	-.384	.075	-.354	.077	-.430
-.452	-.236	.105	-.423	.134	-.385	.133	-.338	.129	-.290	.129	-.247
-.311	-.282	.178	-.364	.209	-.324	.214	-.272	.201	-.243	.209	-.199
-.023	-.252	.286	-.319	.294	-.279	.295	-.246	.294	-.223	.293	-.180
.133	-.193	.396	-.274	.404	-.259	.407	-.227	.397	-.190	.494	-.156
.272	-.155	.514	-.226	.497	-.240	.502	-.223	.495	-.199	.590	-.160
.416	-.126	.618	-.205	.599	-.234	.601	-.217	.594	-.211	.693	-.185
.565	-.094	.733	-.167	.700	-.227	.698	-.240	.693	-.227	.777	-.203
.713	-.081	.835	-.155	.864	-.209	.863	-.233	.784	-.237	.861	-.188
.854	-.089	.919	-.121	.926	-.157	.923	-.181	.856	-.243	.918	-.174
.960	-.082	.987	-.048	.975	-.080	.977	-.067	.926	-.182	.972	-.075
1.074	-.068							.977	-.063		
1.122	-.033										
					LOWER SURFACE						
-.660	.083	-.022	.257	.024	.180	.025	.106	.019	.124	.020	.061
-.616	.032	.038	.033	.075	-.032	.130	-.061	.066	-.095	.076	-.129
-.572	.014	.101	-.046	.297	-.080	.298	-.064	.136	-.082	.136	-.130
-.462	-.024	.185	-.050	.400	-.078	.397	-.057	.214	-.061	.221	-.111
-.329	-.029	.398	-.064	.604	-.064	.501	-.054	.292	-.072	.295	-.105
-.172	-.025	.737	.100	.785	.154	.603	.001	.403	-.054	.396	-.060
-.030	-.071			.967	.164	.703	.057	.489	-.058	.497	-.056
.128	-.113			1.000	.012	.784	.138	.594	-.041	.597	-.019
.418	-.039					.868	.217	.700	.063	.702	.074
.564	.002					.923	.261	.786	.164	.786	.159
.710	.100					.972	.169	.858	.236	.864	.217
.976	.213							.919	.252	.912	.218
1.072	.210							.967	.140		
1.110	.134										
CN=	.3204		.3008		.2915		.2867		.2556		.2090
CM=	.0003		-.0424		-.0606		-.0744		-.0754		-.0695

$\alpha = 2.91^{\circ}$; $C_L = 0.299$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.061	-.021	-.550	.023	-.708	.025	-.605	.022	-.573	.018	-.388
-.567	-.191	.035	-.551	.068	-.549	.079	-.438	.075	-.426	.077	-.479
-.452	-.300	.105	-.433	.134	-.423	.133	-.379	.129	-.339	.129	-.270
-.311	-.293	.178	-.392	.209	-.338	.214	-.307	.201	-.289	.209	-.226
-.023	-.269	.286	-.334	.294	-.306	.295	-.284	.294	-.242	.293	-.198
.133	-.192	.396	-.274	.404	-.273	.407	-.257	.397	-.213	.494	-.185
.272	-.166	.514	-.223	.497	-.253	.502	-.233	.495	-.219	.590	-.181
.416	-.139	.618	-.201	.599	-.228	.601	-.240	.594	-.219	.693	-.195
.565	-.099	.733	-.171	.700	-.229	.698	-.250	.693	-.241	.777	-.211
.713	-.082	.835	-.152	.864	-.195	.863	-.236	.784	-.244	.861	-.195
.854	-.091	.919	-.114	.926	-.158	.923	-.183	.856	-.238	.918	-.171
.980	-.091	.987	-.044	.975	-.071	.977	-.075	.926	-.189	.972	-.075
1.074	-.069							.977	-.059		
1.122	-.042										
					LOWER SURFACE						
-.660	.076	-.022	.287	.024	.226	.025	.163	.019	.155	.020	.177
-.616	.049	.038	.065	.075	.021	.130	-.058	.066	-.029	.076	-.067
-.572	.029	.101	-.018	.297	-.069	.298	-.066	.136	-.057	.136	-.123
-.462	-.019	.185	-.042	.400	-.068	.397	-.058	.214	-.055	.221	-.094
-.329	-.022	.398	-.061	.604	-.061	.501	-.049	.292	-.062	.295	-.096
-.172	-.019	.737	.105	.785	.155	.603	-.004	.403	-.048	.396	-.060
-.030	-.074			.967	.157	.703	.061	.489	-.065	.497	-.061
.128	-.113			1.000	.008	.784	.141	.594	-.047	.597	-.031
.418	-.032					.868	.218	.700	.057	.702	.067
.564	.016					.923	.260	.786	.158	.786	.152
.710	.097					.972	.160	.858	.232	.864	.203
.976	.215							.919	.244	.912	.216
1.072	.211							.967	.139		
1.110	.129										
CN=	.3491		.3214		.3160		.3175		.2874		.2387
CM=	.0099		-.0397		-.0569		-.0737		-.0720		-.0668

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{o}$

(a) M = 0.25. Continued.

$\alpha = 3.43^{o}$; $C_L = 0.343$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.096	-.021	-.682	.023	-.933	.025	-.724	.022	-.661	.018	-.552
-.567	-.218	.035	-.589	.068	-.594	.079	-.505	.075	-.502	.077	-.566
-.452	-.288	.105	-.453	.134	-.461	.133	-.419	.129	-.381	.129	-.326
-.311	-.304	.178	-.426	.209	-.393	.214	-.333	.201	-.323	.209	-.259
-.023	-.275	.286	-.344	.294	-.325	.295	-.292	.294	-.263	.293	-.229
.133	-.195	.396	-.294	.404	-.280	.407	-.274	.397	-.233	.494	-.190
.272	-.169	.514	-.240	.497	-.268	.502	-.251	.495	-.235	.590	-.189
.416	-.134	.618	-.203	.599	-.240	.601	-.250	.594	-.237	.693	-.204
.565	-.106	.733	-.178	.700	-.239	.698	-.252	.693	-.259	.777	-.216
.713	-.084	.835	-.151	.864	-.210	.863	-.241	.784	-.248	.861	-.201
.854	-.089	.919	-.113	.926	-.158	.923	-.174	.856	-.244	.918	-.182
.980	-.096	.987	-.041	.975	-.069	.977	-.070	.926	-.189	.972	-.075
1.074	-.069							.977	-.064		
1.122	-.041										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.085	-.022	.319	.024	.300	.025	.254	.019	.240	.020	.211
-.616	.074	.038	.096	.075	.066	.130	-.020	.066	.027	.076	-.026
-.572	.046	.101	.016	.297	-.052	.298	-.036	.136	-.016	.136	-.070
-.462	-.003	.185	-.011	.400	-.041	.397	-.043	.214	-.037	.221	-.072
-.329	-.006	.398	-.045	.604	-.049	.501	-.035	.292	-.040	.295	-.077
-.172	-.019	.737	.110	.785	.154	.603	.009	.403	-.030	.396	-.058
-.030	-.056			.967	.157	.703	.067	.489	-.051	.497	-.058
.128	-.093			1.000	.005	.784	.141	.594	-.038	.597	-.022
.418	-.024					.868	.229	.700	.068	.702	.075
.564	.028					.923	.262	.786	.159	.786	.152
.710	.108					.972	.163	.858	.234	.864	.208
.976	.220							.919	.241	.912	.208
1.072	.216							.967	.131		
1.110	.130										

	.133	.307	.480	.653	.804	.933
CN=	.3779	.3553	.3635	.3598	.3298	.2787
CM=	.0168	-.0385	-.0559	-.0725	-.0721	-.0646

$\alpha = 3.99^{o}$; $C_L = 0.391$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.104	-.021	-.846	.023	-1.024	.025	-.832	.022	-.827	.018	-.720
-.567	-.251	.035	-.686	.068	-.674	.079	-.580	.075	-.585	.077	-.640
-.452	-.311	.105	-.550	.134	-.525	.133	-.466	.129	-.450	.129	-.354
-.311	-.333	.178	-.445	.209	-.433	.214	-.373	.201	-.349	.209	-.278
-.023	-.291	.286	-.371	.294	-.363	.295	-.319	.294	-.307	.293	-.244
.133	-.216	.396	-.320	.404	-.307	.407	-.290	.397	-.241	.494	-.200
.272	-.190	.514	-.244	.497	-.279	.502	-.264	.495	-.249	.590	-.201
.416	-.142	.618	-.224	.599	-.259	.601	-.243	.594	-.247	.693	-.211
.565	-.117	.733	-.186	.700	-.254	.698	-.271	.693	-.248	.777	-.221
.713	-.054	.835	-.166	.864	-.220	.863	-.236	.784	-.262	.861	-.201
.854	-.102	.919	-.117	.926	-.157	.923	-.188	.856	-.247	.918	-.180
.980	-.092	.987	-.049	.975	-.061	.977	-.055	.926	-.185	.972	-.074
1.074	-.078							.977	-.051		
1.122	-.041										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.090	-.022	.330	.024	.330	.025	.313	.019	.312	.020	.272
-.616	.079	.038	.147	.075	.117	.130	.035	.066	.075	.076	.034
-.572	.050	.101	.054	.297	-.026	.298	-.009	.136	.018	.136	-.040
-.462	.016	.185	.016	.400	-.022	.397	-.022	.214	-.004	.221	-.042
-.329	.011	.398	-.022	.604	-.040	.501	-.024	.292	-.025	.295	-.052
-.172	.000	.737	.106	.785	.168	.603	.022	.403	-.014	.396	-.038
-.030	-.047			.967	.154	.703	.070	.489	-.024	.497	-.045
.128	-.083			1.000	.005	.784	.150	.594	-.025	.597	-.014
.418	-.019					.868	.232	.700	.080	.702	.080
.564	.028					.923	.273	.786	.160	.786	.151
.710	.119					.972	.169	.858	.239	.864	.207
.976	.226							.919	.251	.912	.219
1.072	.223							.967	.138		
1.110	.129										

	.133	.307	.480	.653	.804	.933
CN=	.4236	.4100	.4106	.4047	.3758	.3187
CM=	.0271	-.0357	-.0571	-.0716	-.0706	-.0634

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25. Continued.

$\alpha = 4.97^o$; $C_L = 0.472$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
					UPPER SURFACE						
-.660	-.138	-.021	-1.083	.023	-1.259	.025	-1.047	.022	-1.066	.018	-.957
-.567	-.274	.035	-.766	.068	-.819	.079	-.681	.075	-.674	.077	-.695
-.452	-.386	.105	-.618	.134	-.603	.133	-.584	.129	-.524	.129	-.433
-.311	-.357	.178	-.529	.209	-.484	.214	-.443	.201	-.417	.209	-.338
-.023	-.304	.286	-.422	.294	-.396	.295	-.374	.294	-.344	.293	-.291
.133	-.257	.396	-.353	.404	-.333	.407	-.317	.397	-.300	.494	-.218
.272	-.197	.514	-.278	.497	-.298	.502	-.301	.495	-.284	.590	-.226
.416	-.158	.618	-.244	.599	-.284	.601	-.284	.594	-.277	.693	-.226
.565	-.133	.733	-.173	.700	-.266	.698	-.281	.693	-.270	.777	-.234
.713	-.110	.835	-.157	.864	-.213	.863	-.245	.784	-.270	.861	-.212
.854	-.125	.919	-.106	.926	-.148	.923	-.184	.856	-.243	.918	-.177
.980	-.100	.987	-.036	.975	-.068	.977	-.060	.926	-.176	.972	-.075
1.074	-.080							.977	-.054		
1.122	-.048										
					LOWER SURFACE						
-.660	.091	-.022	.367	.024	.425	.025	.378	.019	.414	.020	.385
-.616	.108	.038	.202	.075	.189	.130	.093	.066	.166	.076	.098
-.572	.081	.101	.125	.297	.013	.298	.006	.136	.077	.136	.019
-.462	.051	.185	.055	.400	.012	.397	.014	.214	.054	.221	-.015
-.329	.033	.398	.003	.604	-.023	.501	-.007	.292	.037	.295	-.015
-.172	.028	.737	.137	.785	.175	.603	.040	.403	.004	.396	-.013
-.030	-.029			.967	.162	.703	.093	.489	-.014	.497	-.034
.128	-.056			1.000	-.002	.784	.157	.594	-.018	.597	-.012
.418	.012					.868	.232	.700	.086	.702	.079
.564	.054					.923	.269	.786	.171	.786	.150
.710	.123					.972	.166	.858	.243	.864	.206
.976	.234							.919	.254	.912	.211
1.072	.230							.967	.136		
1.110	.149										
CN=	.5022		.4688		.4814		.4779		.4503		.3800
CM=	.0432		-.0357		-.0554		-.0720		-.0693		-.0606

$\alpha = 6.03^o$; $C_L = 0.559$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
					UPPER SURFACE						
-.660	-.191	-.021	-1.400	.023	-1.841	.025	-1.466	.022	-1.400	.018	-1.321
-.567	-.323	.035	-.921	.068	-1.057	.079	-.871	.075	-.864	.077	-.957
-.452	-.385	.105	-.703	.134	-.732	.133	-.651	.129	-.649	.129	-.584
-.311	-.401	.178	-.578	.209	-.569	.214	-.510	.201	-.503	.209	-.425
-.023	-.331	.286	-.471	.294	-.466	.295	-.422	.294	-.403	.293	-.334
.133	-.250	.396	-.395	.404	-.384	.407	-.362	.397	-.339	.494	-.238
.272	-.224	.514	-.309	.497	-.326	.502	-.320	.495	-.316	.590	-.241
.416	-.178	.618	-.259	.599	-.289	.601	-.291	.594	-.308	.693	-.242
.565	-.142	.733	-.212	.700	-.271	.698	-.292	.693	-.291	.777	-.239
.713	-.121	.835	-.176	.864	-.203	.863	-.237	.784	-.284	.861	-.212
.854	-.117	.919	-.109	.926	-.133	.923	-.174	.856	-.256	.918	-.177
.980	-.116	.987	-.049	.975	-.055	.977	-.063	.926	-.181	.972	-.075
1.074	-.104							.977	-.056		
1.122	-.063										
					LOWER SURFACE						
-.660	.086	-.022	.369	.024	.496	.025	.458	.019	.465	.020	.486
-.616	.118	.038	.267	.075	.287	.130	.163	.066	.238	.076	.180
-.572	.106	.101	.169	.297	.065	.298	.059	.136	.130	.136	.076
-.462	.074	.185	.104	.400	.050	.357	.044	.214	.091	.221	.030
-.329	.054	.398	.028	.604	-.002	.501	.042	.292	.061	.295	.031
-.172	.048	.737	.131	.785	.183	.603	.053	.403	.045	.396	.016
-.030	.010			.967	.160	.703	.096	.489	.015	.497	-.004
.128	-.032			1.000	-.006	.784	.173	.594	.005	.597	.013
.418	.036					.868	.235	.700	.096	.702	.086
.564	.063					.923	.277	.786	.182	.786	.154
.710	.148					.972	.162	.858	.252	.864	.218
.976	.242							.919	.262	.912	.214
1.072	.244							.967	.137		
1.110	.138										
CN=	.5732		.5678		.5866		.5659		.5395		.4732
CM=	.0565		-.0326		-.0465		-.0671		-.0695		-.0557

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(a) M = 0.25. Continued.

$\alpha = 6.98^{\circ}$; $C_L = 0.638$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.220	-.021	-1.655	.023	-2.075	.025	-1.853	.022	-1.903	.018	-1.633
-.567	-.380	.035	-1.040	.068	-1.173	.079	-1.016	.075	-1.003	.077	-1.019
-.452	-.427	.105	-.796	.134	-.780	.133	-.768	.129	-.742	.129	-.631
-.311	-.430	.178	-.657	.209	-.622	.214	-.569	.201	-.563	.209	-.445
-.023	-.350	.286	-.505	.294	-.495	.295	-.478	.294	-.455	.293	-.363
.133	-.284	.396	-.406	.404	-.413	.407	-.391	.397	-.368	.494	-.266
.272	-.239	.514	-.314	.497	-.356	.502	-.351	.495	-.343	.590	-.252
.416	-.185	.618	-.268	.599	-.312	.601	-.321	.594	-.309	.693	-.252
.565	-.163	.733	-.203	.700	-.283	.698	-.298	.693	-.293	.777	-.252
.713	-.133	.835	-.156	.864	-.210	.863	-.229	.784	-.268	.861	-.216
.854	-.139	.919	-.103	.926	-.131	.923	-.160	.856	-.239	.918	-.181
.980	-.116	.987	-.060	.975	-.067	.977	-.052	.926	-.152	.972	-.079
1.074	-.103							.977	-.058		
1.122	-.056										
					LOWER SURFACE						
-.660	.077	-.022	.345	.024	.509	.025	.518	.019	.501	.020	.485
-.616	.121	.038	.306	.075	.345	.130	.208	.066	.320	.076	.248
-.572	.136	.101	.212	.297	.101	.298	.091	.136	.193	.136	.153
-.462	.057	.185	.132	.400	.064	.397	.076	.214	.127	.221	.076
-.329	.075	.398	.067	.604	.006	.501	.050	.292	.100	.295	.053
-.172	.073	.737	.141	.785	.186	.603	.078	.403	.064	.396	.022
-.030	.024			.967	.153	.703	.121	.489	.032	.497	.000
.128	-.005			1.000	-.022	.784	.171	.594	.025	.597	.021
.418	.043					.868	.248	.700	.111	.702	.082
.564	.068					.923	.281	.786	.180	.786	.151
.710	.166					.972	.164	.858	.249	.864	.199
.976	.271							.919	.257	.912	.199
1.072	.241							.967	.126		
1.110	.145										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.6466	.6299	.6427	.6445	.6121	.5219
CM=	.0700	-.0282	-.0453	-.0633	-.0600	-.0513

$\alpha = 8.00^{\circ}$; $C_L = 0.714$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.281	-.021	-2.040	.023	-2.482	.025	-1.614	.022	-2.306	.018	-2.004
-.567	-.416	.035	-1.170	.068	-1.319	.079	-1.143	.075	-1.127	.077	-1.130
-.452	-.475	.105	-.850	.134	-.893	.133	-.852	.129	-.822	.129	-.718
-.311	-.462	.178	-.682	.209	-.665	.214	-.649	.201	-.634	.209	-.519
-.023	-.373	.286	-.545	.294	-.545	.295	-.531	.294	-.485	.293	-.411
.133	-.296	.396	-.419	.404	-.442	.407	-.429	.397	-.405	.494	-.291
.272	-.248	.514	-.330	.497	-.377	.502	-.371	.495	-.362	.590	-.272
.416	-.209	.618	-.277	.599	-.324	.601	-.339	.594	-.320	.693	-.262
.565	-.163	.733	-.199	.700	-.280	.698	-.311	.693	-.299	.777	-.248
.713	-.155	.835	-.160	.864	-.188	.863	-.240	.784	-.268	.861	-.216
.854	-.143	.919	-.095	.926	-.117	.923	-.157	.856	-.218	.918	-.167
.980	-.138	.987	-.048	.975	-.058	.977	-.060	.926	-.140	.972	-.076
1.074	-.099							.977	-.049		
1.122	-.062										
					LOWER SURFACE						
-.660	.074	-.022	.301	.024	.516	.025	.540	.019	.515	.020	.502
-.616	.149	.038	.346	.075	.372	.130	.269	.066	.373	.076	.300
-.572	.158	.101	.248	.297	.137	.298	.125	.136	.237	.136	.174
-.462	.131	.185	.183	.400	.101	.397	.097	.214	.158	.221	.077
-.329	.111	.398	.087	.604	.027	.501	.081	.292	.138	.295	.075
-.172	.095	.737	.148	.785	.190	.603	.099	.403	.089	.396	.036
-.030	.061			.967	.148	.703	.119	.489	.053	.497	.008
.128	.022			1.000	-.031	.784	.165	.594	.041	.597	.025
.418	.070					.868	.245	.700	.105	.702	.084
.564	.101					.923	.283	.786	.181	.786	.144
.710	.178					.972	.155	.858	.250	.864	.195
.976	.272							.919	.251	.912	.190
1.072	.260							.967	.119		
1.110	.139										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.7224	.6920	.7074	.6877	.6749	.5781
CM=	.0895	-.0225	-.0395	-.0675	-.0533	-.0454

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; δ_h = -2.5°

(a) M = 0.25. Continued.

$\alpha = 9.02^{\circ}$; C_L = 0.791

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
				UPPER SURFACE							
-.660	-.325	-.021	-2.368	.023	-2.590	.025	-1.728	.022	-2.912	.018	-2.459
-.567	-.444	.035	-1.312	.068	-1.508	.079	-1.280	.075	-1.322	.077	-1.228
-.452	-.499	.105	-.966	.134	-.966	.133	-.941	.129	-.936	.129	-.812
-.311	-.508	.178	-.754	.209	-.728	.214	-.699	.201	-.692	.209	-.586
-.023	-.400	.286	-.596	.294	-.580	.295	-.574	.294	-.553	.293	-.462
.133	-.318	.396	-.460	.404	-.452	.407	-.465	.397	-.445	.494	-.308
.272	-.270	.514	-.347	.497	-.380	.502	-.417	.495	-.396	.590	-.288
.416	-.223	.618	-.281	.599	-.319	.601	-.363	.594	-.344	.693	-.266
.565	-.180	.733	-.209	.700	-.278	.698	-.327	.693	-.304	.777	-.251
.713	-.157	.835	-.153	.864	-.168	.863	-.228	.784	-.267	.861	-.208
.854	-.150	.919	-.093	.926	-.104	.923	-.145	.856	-.210	.918	-.161
.980	-.130	.987	-.056	.975	-.072	.977	-.063	.926	-.127	.972	-.084
1.074	-.097							.977	-.059		
1.122	-.054										
				LOWER SURFACE							
-.660	.068	-.022	.233	.024	.512	.025	.525	.019	.494	.020	.503
-.616	.172	.038	.370	.075	.397	.130	.313	.066	.413	.076	.357
-.572	.163	.101	.291	.297	.170	.298	.161	.136	.288	.136	.221
-.462	.154	.165	.226	.400	.131	.397	.130	.214	.221	.221	.137
-.329	.130	.398	.119	.604	.028	.501	.085	.292	.166	.295	.091
-.172	.119	.737	.160	.785	.194	.603	.107	.403	.118	.396	.055
-.030	.062			.967	.141	.703	.128	.489	.077	.497	.032
.128	.032			1.000	-.053	.784	.175	.594	.054	.597	.033
.418	.096					.868	.260	.700	.121	.702	.091
.564	.127					.923	.296	.786	.199	.786	.144
.710	.203					.972	.166	.858	.253	.864	.191
.976	.280							.919	.256	.912	.189
1.072	.268							.967	.122		
1.110	.168										
CN=	.7887		.7661		.7485		.7445		.7657		.6469
CM=	.1022		-.0200		-.0348		-.0695		-.0485		-.0401

$\alpha = 10.02^{\circ}$; C_L = 0.868

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
				UPPER SURFACE							
-.660	-.390	-.021	-2.804	.023	-2.733	.025	-2.029	.022	-3.134	.018	-.936
-.567	-.531	.035	-1.506	.068	-1.641	.079	-1.434	.075	-2.010	.077	-.940
-.452	-.563	.105	-1.065	.134	-1.014	.133	-1.058	.129	-1.054	.129	-.921
-.311	-.536	.178	-.853	.209	-.806	.214	-.786	.201	-.775	.209	-.764
-.023	-.422	.286	-.634	.294	-.635	.295	-.618	.294	-.595	.293	-.783
.133	-.365	.396	-.507	.404	-.495	.407	-.508	.397	-.454	.494	-.554
.272	-.289	.514	-.377	.497	-.411	.502	-.437	.495	-.413	.590	-.448
.416	-.253	.618	-.306	.599	-.339	.601	-.371	.594	-.343	.693	-.376
.565	-.193	.733	-.229	.700	-.285	.698	-.330	.693	-.295	.777	-.319
.713	-.166	.835	-.174	.864	-.173	.863	-.214	.784	-.252	.861	-.234
.854	-.170	.919	-.107	.926	-.115	.923	-.146	.856	-.181	.918	-.207
.980	-.161	.987	-.082	.975	-.084	.977	-.071	.926	-.112	.972	-.132
1.074	-.118							.977	-.073		
1.122	-.057										
				LOWER SURFACE							
-.660	.053	-.022	.140	.024	.475	.025	.518	.019	.474	.020	.479
-.616	.169	.038	.394	.075	.453	.130	.358	.066	.441	.076	.391
-.572	.195	.101	.319	.297	.179	.298	.201	.136	.340	.136	.252
-.462	.168	.165	.247	.400	.132	.397	.149	.214	.248	.221	.185
-.329	.148	.398	.132	.604	.051	.501	.106	.292	.191	.295	.121
-.172	.132	.737	.167	.785	.188	.603	.132	.403	.140	.396	.067
-.030	.104			.967	.127	.703	.128	.489	.105	.497	.039
.128	.054			1.000	-.079	.784	.176	.594	.069	.597	.034
.418	.119					.868	.265	.700	.132	.702	.082
.564	.140					.923	.302	.786	.191	.786	.136
.710	.216					.972	.156	.858	.257	.864	.185
.976	.283							.919	.257	.912	.194
1.072	.234							.967	.117		
1.110	.165										
CN=	.8747		.8492		.7985		.8120		.8459		.7114
CM=	.1216		-.0155		-.0343		-.0661		-.0378		-.0864

[redacted]

TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25. Concluded.

$\alpha = 11.03^o$; $C_L = 0.946$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
UPPER SURFACE												
	-.660	-.470	-.021	-3.387	.023	-2.822	.025	-2.321	.022	-2.176	.018	-.798
	-.567	-.568	.035	-1.657	.068	-1.805	.079	-1.600	.075	-2.064	.077	-.737
	-.452	-.599	.105	-1.137	.134	-1.166	.133	-1.165	.129	-1.755	.129	-.720
	-.311	-.561	.178	-.913	.209	-.912	.214	-.859	.201	-1.757	.209	-.690
	-.023	-.446	.286	-.684	.294	-.691	.295	-.670	.294	-.920	.293	-.616
	.133	-.369	.396	-.537	.404	-.533	.407	-.530	.397	-.342	.494	-.540
	.272	-.315	.514	-.400	.497	-.435	.502	-.449	.495	-.291	.590	-.467
	.416	-.298	.618	-.332	.599	-.345	.601	-.366	.594	-.260	.693	-.441
	.565	-.219	.733	-.245	.700	-.284	.698	-.331	.693	-.261	.777	-.388
	.713	-.202	.835	-.186	.864	-.155	.863	-.199	.784	-.228	.861	-.339
	.854	-.194	.919	-.134	.926	-.123	.923	-.134	.856	-.223	.918	-.325
	.980	-.163	.987	-.088	.975	-.104	.977	-.092	.926	-.172	.972	-.293
	1.074	-.106							.977	-.071		
	1.122	-.097										
LOWER SURFACE												
	-.660	.022	-.022	.043	.024	.441	.025	.493	.019	.398	.020	.508
	-.616	.170	.038	.394	.075	.472	.130	.393	.066	.497	.076	.404
	-.572	.200	.101	.356	.297	.221	.298	.227	.136	.369	.136	.276
	-.462	.184	.185	.294	.400	.160	.397	.177	.214	.248	.221	.184
	-.329	.178	.398	.169	.604	.056	.501	.132	.292	.218	.295	.129
	-.172	.156	.737	.173	.785	.197	.603	.142	.403	.154	.396	.088
	-.030	.125			.967	.107	.703	.135	.489	.116	.497	.045
	.128	.094			1.000	-.111	.784	.175	.594	.078	.597	.034
	.418	.130					.868	.261	.700	.134	.702	.083
	.564	.166					.923	.292	.786	.194	.786	.141
	.710	.222					.972	.143	.858	.253	.864	.174
	.976	.298							.919	.255	.912	.187
	1.072	.239							.967	.108		
	1.110	.173										
CN=		.9446		.9368		.8546		.8684		.9366		.6909
CM=		.1346		-.0175		-.0319		-.0608		-.0346		-.1110

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50.

$\alpha = -5.10^{\circ}$; $C_L = -0.431$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	.033	-.021	.344	.023	.415	.025	.423	.022	.500	.018	.517
-.567	.011	.035	.187	.068	.270	.079	.277	.075	.316	.077	.239
-.452	-.051	.105	.127	.134	.158	.133	.195	.129	.235	.129	.210
-.311	-.088	.178	.054	.209	.102	.214	.133	.201	.159	.209	.139
-.023	-.065	.286	.018	.294	.060	.295	.099	.294	.115	.293	.085
.133	-.024	.396	.002	.404	.023	.407	.044	.397	.070	.494	-.005
.272	.003	.514	-.005	.497	-.006	.502	.011	.495	.020	.590	-.045
.416	.015	.618	-.031	.599	-.033	.601	-.032	.594	-.026	.693	-.090
.565	.036	.733	-.041	.700	-.073	.658	-.085	.693	-.075	.777	-.117
.713	.032	.835	-.066	.864	-.127	.863	-.150	.784	-.120	.861	-.139
.854	.002	.919	-.072	.926	-.116	.923	-.140	.856	-.160	.918	-.153
.980	-.020	.987	-.038	.975	-.055	.977	-.058	.926	-.158	.972	-.090
1.074	-.030							.977	-.069		
1.122	-.019										
LOWER SURFACE											
-.660	-.069	-.022	-1.050	.024	-1.808	.025	-2.425	.019	-1.501	.020	-.703
-.616	-.215	.036	-.781	.075	-.993	.130	-.801	.066	-1.201	.076	-.682
-.572	-.242	.101	-.672	.257	-.444	.258	-.444	.136	-1.152	.136	-.601
-.462	-.260	.185	-.516	.400	-.352	.357	-.358	.214	-.934	.221	-.551
-.329	-.220	.358	-.322	.604	-.206	.501	-.261	.292	-.564	.295	-.487
-.172	-.212	.737	.002	.785	.048	.603	-.112	.403	-.267	.396	-.375
-.030	-.275			.967	.155	.703	.000	.489	-.225	.497	-.330
.128	-.315			1.000	.051	.784	.055	.594	-.146	.597	-.275
.418	-.227					.868	.117	.700	-.034	.702	-.228
.564	-.151					.923	.135	.786	.058	.786	-.161
.710	-.027					.972	.105	.858	.127	.864	-.115
.976	.176							.919	.141	.912	-.109
1.072	.170							.967	.114		
1.110	.134										
CN=	-.2343		-.3621		-.3859		-.4370		-.4303		-.3851
CM=	-.1031		-.0458		-.0635		-.0837		-.0732		-.0017

$\alpha = -4.08^{\circ}$; $C_L = -0.339$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	.035	-.021	.299	.023	.360	.025	.373	.022	.459	.018	.499
-.567	-.007	.035	.124	.068	.184	.079	.217	.075	.253	.077	.197
-.452	-.076	.105	.060	.134	.099	.133	.147	.129	.185	.129	.182
-.311	-.119	.178	.013	.209	.046	.214	.092	.201	.120	.209	.117
-.023	-.099	.286	-.023	.294	.018	.295	.049	.294	.081	.293	.059
.133	-.041	.396	-.043	.404	-.010	.407	.014	.397	.040	.494	-.016
.272	-.021	.514	-.032	.497	-.044	.502	-.029	.495	-.003	.590	-.055
.416	.000	.618	-.060	.599	-.065	.601	-.056	.594	-.046	.693	-.096
.565	.006	.733	-.065	.700	-.105	.658	-.100	.693	-.088	.777	-.133
.713	.012	.835	-.091	.864	-.144	.863	-.167	.784	-.133	.861	-.133
.854	-.016	.919	-.080	.926	-.125	.923	-.147	.856	-.158	.918	-.140
.980	-.026	.987	-.048	.975	-.057	.977	-.057	.926	-.153	.972	-.061
1.074	-.041							.977	-.042		
1.122	-.023										
LOWER SURFACE											
-.660	-.068	-.022	-.827	.024	-1.292	.025	-1.911	.019	-2.183	.020	-1.642
-.616	-.164	.038	-.677	.075	-.896	.130	-.692	.066	-1.082	.076	-1.157
-.572	-.210	.101	-.585	.257	-.386	.258	-.384	.136	-.695	.136	-.936
-.462	-.217	.185	-.469	.400	-.310	.357	-.306	.214	-.485	.221	-.472
-.329	-.207	.398	-.295	.604	-.191	.501	-.236	.292	-.389	.295	-.394
-.172	-.175	.737	.025	.785	.065	.603	-.096	.403	-.308	.396	-.267
-.030	-.249			.967	.170	.703	.013	.489	-.262	.497	-.174
.128	-.292			1.000	.043	.784	.059	.594	-.171	.597	-.090
.418	-.202					.868	.142	.700	-.030	.702	.006
.564	-.138					.923	.158	.786	.048	.786	.049
.710	-.015					.972	.122	.858	.089	.864	.075
.976	.175							.919	.145	.912	.089
1.072	.172							.967	.114		
1.110	.135										
CN=	-.1590		-.2758		-.2867		-.3294		-.3574		-.3363
CM=	-.0684		-.0457		-.0649		-.0819		-.0702		-.0720

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TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(b) M = 0.50. Continued.

$\alpha = -3.08^\circ$; $C_L = -0.247$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.033	-.021	.253	.023	.274	.025	.280	.022	.394	.018	.463
-.567	-.026	.035	.058	.068	.132	.079	.131	.075	.179	.077	.141
-.452	-.108	.105	-.016	.134	.034	.133	.089	.129	.125	.129	.133
-.311	-.134	.178	-.046	.209	.008	.214	.035	.201	.075	.209	.073
-.023	-.125	.286	-.073	.294	-.026	.295	.008	.294	.040	.293	.036
.133	-.065	.396	-.080	.404	-.050	.407	-.028	.397	.007	.494	-.038
.272	-.047	.514	-.068	.497	-.072	.502	-.048	.495	-.037	.590	-.071
.416	-.021	.618	-.090	.599	-.085	.601	-.085	.594	-.073	.693	-.105
.565	-.007	.733	-.081	.700	-.117	.698	-.125	.693	-.113	.777	-.134
.713	-.004	.835	-.097	.864	-.162	.863	-.182	.784	-.151	.861	-.136
.854	-.026	.919	-.094	.926	-.132	.923	-.150	.856	-.171	.918	-.132
.980	-.035	.987	-.049	.975	-.057	.977	-.050	.926	-.155	.972	-.040
1.074	-.039							.977	-.045		
1.122	-.024										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.029	-.022	-.611	.024	-1.040	.025	-1.332	.019	-1.672	.020	-1.825
-.616	-.133	.038	-.550	.075	-.728	.130	-.559	.066	-.874	.076	-.893
-.572	-.164	.101	-.465	.297	-.345	.298	-.343	.136	-.606	.136	-.611
-.462	-.187	.185	-.395	.400	-.288	.397	-.284	.214	-.435	.221	-.414
-.329	-.168	.398	-.256	.604	-.174	.501	-.211	.292	-.332	.295	-.311
-.172	-.161	.737	.043	.785	.077	.603	-.089	.403	-.267	.396	-.234
-.030	-.217			.967	.171	.703	.027	.489	-.233	.497	-.162
.128	-.272			1.000	.041	.784	.078	.594	-.163	.597	-.070
.418	-.177					.868	.155	.700	-.008	.702	.037
.564	-.110					.923	.175	.786	.080	.786	.076
.710	.005					.972	.133	.858	.130	.864	.104
.976	.180							.919	.164	.912	.097
1.072	.175							.967	.132		
1.110	.126										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	-.0859	-.1769	-.2046	-.2152	-.2574	-.2606
CM=	-.0742	-.0511	-.0640	-.0766	-.0723	-.0741

$\alpha = -2.08^\circ$; $C_L = -0.154$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.034	-.021	.170	.023	.141	.025	.188	.022	.311	.018	.387
-.567	-.036	.035	-.026	.068	.025	.079	.074	.075	.113	.077	.041
-.452	-.129	.105	-.073	.134	-.026	.133	.010	.129	.052	.129	.084
-.311	-.167	.178	-.101	.209	-.070	.214	-.015	.201	.015	.209	.033
-.023	-.143	.286	-.120	.294	-.069	.295	-.037	.294	-.008	.293	-.013
.133	-.089	.396	-.112	.404	-.095	.407	-.066	.397	-.050	.494	-.069
.272	-.072	.514	-.095	.497	-.102	.502	-.089	.495	-.072	.590	-.085
.416	-.038	.618	-.110	.599	-.122	.601	-.117	.594	-.104	.693	-.128
.565	-.022	.733	-.101	.700	-.142	.698	-.153	.693	-.141	.777	-.150
.713	-.016	.835	-.110	.864	-.176	.863	-.196	.784	-.172	.861	-.144
.854	-.035	.919	-.101	.926	-.141	.923	-.163	.856	-.192	.918	-.140
.980	-.049	.987	-.052	.975	-.066	.977	-.054	.926	-.165	.972	-.045
1.074	-.047							.977	-.047		
1.122	-.021										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.010	-.022	-.355	.024	-.706	.025	-.884	.019	-1.307	.020	-1.364
-.616	-.110	.038	-.453	.075	-.572	.130	-.470	.066	-.744	.076	-.762
-.572	-.130	.101	-.420	.297	-.295	.298	-.301	.136	-.500	.136	-.528
-.462	-.154	.185	-.337	.400	-.249	.397	-.240	.214	-.352	.221	-.359
-.329	-.150	.398	-.233	.604	-.154	.501	-.191	.292	-.287	.295	-.266
-.172	-.138	.737	.048	.785	.098	.603	-.078	.403	-.251	.396	-.211
-.030	-.207			.967	.168	.703	.038	.489	-.209	.497	-.153
.128	-.239			1.000	.033	.784	.099	.594	-.145	.597	-.069
.418	-.159					.868	.153	.700	.004	.702	.055
.564	-.099					.923	.177	.786	.101	.786	.119
.710	.023					.972	.135	.858	.154	.864	.155
.976	.186							.919	.181	.912	.173
1.072	.178							.967	.135		
1.110	.126										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	-.0196	-.0555	-.1022	-.1142	-.1637	-.1630
CM=	-.0625	-.0485	-.0640	-.0745	-.0750	-.0799

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = -1.08^{\circ}$; $C_L = -0.578$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.024	-.021	.067	.023	-.011	.025	.074	.022	.203	.018	.311
-.567	-.070	.035	-.120	.068	-.077	.079	-.021	.075	.003	.077	-.061
-.452	-.155	.105	-.137	.134	-.132	.133	-.071	.129	-.012	.129	.009
-.311	-.197	.178	-.169	.209	-.121	.214	-.063	.201	-.039	.209	-.026
-.023	-.181	.286	-.159	.294	-.112	.295	-.079	.294	-.046	.293	-.048
.133	-.124	.396	-.154	.404	-.131	.407	-.098	.397	-.068	.494	-.085
.272	-.092	.514	-.135	.497	-.139	.502	-.123	.495	-.103	.590	-.108
.416	-.068	.618	-.125	.599	-.153	.601	-.142	.594	-.127	.693	-.144
.565	-.045	.733	-.120	.700	-.173	.698	-.175	.693	-.166	.777	-.168
.713	-.038	.835	-.128	.864	-.187	.863	-.212	.784	-.183	.861	-.148
.854	-.052	.919	-.107	.926	-.154	.923	-.160	.856	-.207	.918	-.148
.960	-.056	.987	-.052	.975	-.073	.977	-.052	.926	-.173	.972	-.049
1.074	-.055							.977	-.046		
1.122	-.033										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.025	-.022	-.194	.024	-.475	.025	-.577	.019	-.852	.020	-.752
-.616	-.064	.038	-.304	.075	-.456	.130	-.361	.066	-.668	.076	-.622
-.572	-.089	.101	-.304	.297	-.229	.298	-.247	.136	-.382	.136	-.496
-.462	-.127	.185	-.266	.400	-.209	.397	-.202	.214	-.306	.221	-.281
-.329	-.113	.398	-.197	.604	-.137	.501	-.164	.292	-.243	.295	-.224
-.172	-.105	.737	.067	.785	.121	.603	-.068	.403	-.195	.396	-.178
-.030	-.174			.967	.166	.703	.038	.489	-.176	.497	-.133
.128	-.215			1.000	.027	.784	.117	.594	-.131	.597	-.063
.418	-.132					.868	.165	.700	.016	.702	.063
.564	-.074					.923	.187	.786	.127	.786	.141
.710	.038					.972	.143	.858	.183	.864	.198
.976	.192							.919	.205	.912	.194
1.072	.172							.967	.144		
1.110	.130										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.0693	-.0005	-.0060	-.0198	-.0712	-.0671
CM=	-.0464	-.0491	-.0659	-.0730	-.0764	-.0782

$\alpha = 0^{\circ}$; $C_L = 0.394$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.006	-.021	-.082	.023	-.142	.025	-.092	.022	.043	.018	.153
-.567	-.108	.035	-.249	.068	-.188	.079	-.143	.075	-.095	.077	-.152
-.452	-.193	.105	-.230	.134	-.184	.133	-.138	.129	-.096	.129	-.065
-.311	-.235	.178	-.239	.209	-.195	.214	-.145	.201	-.116	.209	-.072
-.023	-.198	.286	-.220	.294	-.183	.295	-.140	.294	-.108	.293	-.097
.133	-.146	.396	-.196	.404	-.170	.407	-.149	.397	-.113	.494	-.112
.272	-.114	.514	-.154	.497	-.169	.502	-.154	.495	-.137	.590	-.134
.416	-.080	.618	-.163	.599	-.179	.601	-.171	.594	-.159	.693	-.163
.565	-.057	.733	-.142	.700	-.191	.698	-.198	.693	-.183	.777	-.182
.713	-.050	.835	-.139	.864	-.201	.863	-.225	.784	-.211	.861	-.172
.854	-.060	.919	-.119	.926	-.154	.923	-.169	.856	-.216	.918	-.162
.980	-.060	.987	-.049	.975	-.068	.977	-.050	.926	-.184	.972	-.058
1.074	-.061							.977	-.057		
1.122	-.032										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.055	-.022	-.025	.024	-.226	.025	-.287	.019	-.561	.020	-.523
-.616	-.036	.038	-.209	.075	-.316	.130	-.276	.066	-.458	.076	-.459
-.572	-.061	.101	-.217	.297	-.212	.298	-.187	.136	-.343	.136	-.371
-.462	-.095	.165	-.202	.400	-.155	.397	-.176	.214	-.227	.221	-.225
-.329	-.095	.398	-.151	.604	-.113	.501	-.140	.292	-.186	.295	-.184
-.172	-.091	.737	.076	.785	.135	.603	-.052	.403	-.163	.396	-.150
-.030	-.157			.967	.163	.703	.041	.489	-.149	.497	-.114
.128	-.190			1.000	.019	.784	.127	.594	-.114	.597	-.058
.418	-.107					.868	.181	.700	.034	.702	.065
.564	-.054					.923	.208	.786	.138	.786	.151
.710	.052					.972	.155	.858	.207	.864	.209
.976	.200							.919	.216	.912	.213
1.072	.175							.967	.143		
1.110	.132										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.1399	.0990	.0844	.0808	.0278	.0203
CM=	-.0292	-.0479	-.0653	-.0726	-.0776	-.0786

[redacted]

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = 1.03^{\circ}$; $C_L = 0.130$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .48C X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.012	-.021	-.204	.023	-.344	.025	-.243	.022	-.094	.018	.007
-.567	-.129	.035	-.351	.068	-.312	.079	-.241	.075	-.197	.077	-.272
-.452	-.217	.105	-.305	.134	-.276	.133	-.217	.129	-.172	.129	-.138
-.311	-.256	.178	-.312	.209	-.263	.214	-.199	.201	-.178	.209	-.127
-.023	-.225	.286	-.256	.254	-.214	.295	-.184	.294	-.167	.293	-.135
.133	-.170	.396	-.221	.404	-.195	.407	-.185	.397	-.149	.494	-.128
.272	-.141	.514	-.188	.497	-.207	.502	-.190	.495	-.174	.590	-.149
.416	-.098	.618	-.183	.599	-.201	.601	-.199	.594	-.182	.693	-.175
.565	-.083	.733	-.156	.700	-.212	.658	-.218	.693	-.211	.777	-.199
.713	-.059	.835	-.154	.864	-.206	.863	-.227	.784	-.226	.861	-.188
.854	-.075	.919	-.117	.926	-.162	.923	-.180	.856	-.235	.918	-.169
.980	-.074	.987	-.049	.975	-.072	.977	-.058	.926	-.192	.972	-.068
1.074	-.064							.977	-.058		
1.122	-.034										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .48C X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.059	-.022	.096	.024	-.061	.025	-.137	.019	-.270	.020	-.276
-.616	-.005	.038	-.096	.075	-.204	.130	-.214	.066	-.283	.076	-.317
-.572	-.038	.101	-.146	.297	-.158	.298	-.143	.136	-.230	.136	-.274
-.462	-.072	.185	-.160	.400	-.126	.397	-.137	.214	-.198	.221	-.209
-.329	-.076	.398	-.124	.604	-.102	.501	-.111	.292	-.157	.295	-.154
-.172	-.065	.737	.082	.785	.144	.603	-.037	.403	-.114	.396	-.130
-.030	-.122			.967	.165	.703	.044	.489	-.121	.497	-.092
.128	-.157			1.000	.017	.784	.133	.594	-.093	.597	-.044
.418	-.084					.868	.206	.700	.046	.702	.071
.564	-.033					.923	.232	.786	.152	.786	.158
.710	.066					.972	.161	.858	.230	.864	.215
.976	.208							.919	.242	.912	.230
1.072	.179							.967	.143		
1.110	.128										

	STA .133	STA .307	STA .48C	STA .653	STA .804	STA .933
CN=	.2129	.1752	.1685	.1605	.1219	.0970
CM=	-.0192	-.0455	-.0638	-.0749	-.0793	-.0774

$\alpha = 2.02^{\circ}$; $C_L = 0.220$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .48C X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.034	-.021	-.413	.023	-.574	.025	-.410	.022	-.301	.018	-.182
-.567	-.163	.035	-.433	.068	-.446	.079	-.347	.075	-.323	.077	-.384
-.452	-.261	.105	-.382	.134	-.346	.133	-.316	.129	-.269	.129	-.215
-.311	-.287	.178	-.353	.209	-.312	.214	-.260	.201	-.229	.209	-.175
-.023	-.246	.286	-.300	.294	-.270	.295	-.226	.294	-.199	.293	-.162
.133	-.189	.396	-.271	.404	-.231	.407	-.218	.397	-.184	.494	-.154
.272	-.154	.514	-.211	.497	-.235	.502	-.222	.495	-.201	.590	-.176
.416	-.118	.618	-.208	.599	-.220	.601	-.224	.594	-.207	.693	-.189
.565	-.095	.733	-.165	.700	-.225	.658	-.237	.693	-.230	.777	-.208
.713	-.076	.835	-.156	.864	-.216	.863	-.244	.784	-.248	.861	-.193
.854	-.086	.919	-.118	.926	-.157	.923	-.182	.856	-.248	.918	-.176
.980	-.091	.987	-.044	.975	-.068	.977	-.062	.926	-.188	.972	-.066
1.074	-.067							.977	-.055		
1.122	-.032										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .48C X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.071	-.022	.201	.024	.110	.025	.008	.019	-.022	.020	-.026
-.616	.027	.038	-.000	.075	-.082	.130	-.120	.066	-.159	.076	-.223
-.572	.006	.101	-.064	.297	-.126	.298	-.101	.136	-.137	.136	-.209
-.462	-.033	.185	-.091	.400	-.090	.397	-.101	.214	-.117	.221	-.147
-.329	-.043	.398	-.092	.604	-.084	.501	-.086	.292	-.113	.295	-.137
-.172	-.049	.737	.099	.785	.155	.603	-.016	.403	-.085	.396	-.093
-.030	-.093			.967	.167	.703	.055	.489	-.094	.497	-.078
.128	-.137			1.000	.010	.784	.137	.594	-.075	.597	-.038
.418	-.061					.868	.209	.700	.053	.702	.076
.564	-.014					.923	.255	.786	.158	.786	.162
.710	.066					.972	.165	.858	.237	.864	.220
.976	.223							.919	.253	.912	.228
1.072	.202							.967	.146		
1.110	.139										

	STA .133	STA .307	STA .48C	STA .653	STA .804	STA .933
CN=	.2924	.2633	.2526	.2447	.2099	.1697
CM=	-.0045	-.0443	-.0614	-.0757	-.0775	-.0742

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = 2.49^{\circ}$; $C_L = 0.259$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.051	-.021	-.521	.023	-.662	.025	-.523	.022	-.450	.018	-.323
-.567	-.187	.035	-.516	.068	-.505	.079	-.440	.075	-.391	.077	-.427
-.452	-.264	.105	-.427	.134	-.391	.133	-.344	.129	-.312	.129	-.241
-.311	-.294	.178	-.379	.209	-.338	.214	-.293	.201	-.271	.209	-.206
-.023	-.261	.266	-.323	.294	-.280	.295	-.251	.294	-.227	.293	-.188
.133	-.187	.396	-.282	.404	-.255	.407	-.243	.397	-.202	.494	-.167
.272	-.160	.514	-.221	.497	-.238	.502	-.230	.495	-.214	.590	-.176
.416	-.130	.618	-.209	.599	-.234	.601	-.232	.594	-.219	.693	-.200
.565	-.099	.733	-.176	.700	-.236	.698	-.251	.693	-.238	.777	-.215
.713	-.085	.835	-.157	.864	-.216	.863	-.244	.784	-.246	.861	-.199
.854	-.096	.919	-.117	.926	-.156	.923	-.185	.856	-.245	.918	-.177
.980	-.092	.987	-.039	.975	-.068	.977	-.065	.926	-.185	.972	-.064
1.074	-.076							.977	-.053		
1.122	-.041										
					LOWER SURFACE						
-.660	.065	-.022	.235	.024	.153	.025	.093	.019	.057	.020	.021
-.616	.033	.038	.022	.075	-.055	.130	-.090	.066	-.112	.076	-.146
-.572	.013	.101	-.045	.297	-.094	.298	-.086	.136	-.087	.136	-.156
-.462	-.037	.185	-.074	.400	-.084	.397	-.081	.214	-.106	.221	-.140
-.329	-.043	.398	-.085	.604	-.081	.501	-.075	.292	-.098	.295	-.113
-.172	-.039	.737	.101	.785	.156	.603	-.011	.403	-.071	.396	-.079
-.030	-.095			.967	.162	.703	.058	.489	-.089	.497	-.072
.128	-.127			1.000	.011	.784	.142	.594	-.058	.597	-.030
.418	-.056					.868	.216	.700	.060	.702	.077
.564	-.006					.923	.259	.786	.165	.786	.161
.710	.068					.972	.160	.858	.240	.864	.219
.976	.223							.919	.257	.912	.223
1.072	.196							.967	.146		
1.110	.138										
CN=	.3131		.2953		.2839		.2866		.2512		.2060
CM=	-.0021		-.0417		-.0603		-.0748		-.0759		-.0721

$\alpha = 2.98^{\circ}$; $C_L = 0.305$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.076	-.021	-.579	.023	-.839	.025	-.628	.022	-.545	.018	-.412
-.567	-.211	.035	-.556	.068	-.546	.079	-.486	.075	-.444	.077	-.538
-.452	-.283	.105	-.462	.134	-.429	.133	-.382	.129	-.345	.129	-.286
-.311	-.316	.178	-.421	.209	-.381	.214	-.318	.201	-.291	.209	-.223
-.023	-.282	.286	-.353	.294	-.317	.295	-.284	.294	-.273	.293	-.212
.133	-.211	.396	-.296	.404	-.279	.407	-.262	.397	-.225	.494	-.177
.272	-.163	.514	-.235	.497	-.261	.502	-.246	.495	-.231	.590	-.190
.416	-.143	.618	-.210	.599	-.249	.601	-.245	.594	-.227	.693	-.210
.565	-.102	.733	-.182	.700	-.245	.698	-.262	.693	-.242	.777	-.223
.713	-.091	.835	-.159	.864	-.223	.863	-.249	.784	-.250	.861	-.200
.854	-.101	.919	-.119	.926	-.156	.923	-.184	.856	-.248	.918	-.184
.980	-.092	.987	-.041	.975	-.064	.977	-.062	.926	-.188	.972	-.064
1.074	-.076							.977	-.058		
1.122	-.042										
					LOWER SURFACE						
-.660	.085	-.022	.295	.024	.222	.025	.156	.019	.161	.020	.133
-.616	.045	.038	.062	.075	.006	.130	-.044	.066	-.046	.076	-.127
-.572	.020	.101	-.012	.297	-.068	.298	-.057	.136	-.086	.136	-.142
-.462	-.014	.185	-.039	.400	-.068	.397	-.063	.214	-.054	.221	-.106
-.329	-.015	.398	-.063	.604	-.061	.501	-.060	.292	-.074	.295	-.102
-.172	-.012	.737	.110	.785	.162	.603	.001	.403	-.055	.396	-.068
-.030	-.072			.967	.165	.703	.064	.489	-.069	.497	-.067
.128	-.117			1.000	.010	.784	.145	.594	-.050	.597	-.025
.418	-.039					.868	.224	.700	.066	.702	.076
.564	.005					.923	.262	.786	.166	.786	.159
.710	.100					.972	.167	.858	.239	.864	.223
.976	.232							.919	.258	.912	.225
1.072	.195							.967	.141		
1.110	.144										
CN=	.3650		.3398		.3358		.3304		.2925		.2428
CM=	.0114		-.0420		-.0608		-.0755		-.0753		-.0707

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = 3.48^{\circ}$; $C_L = 0.351$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.068	-.021	-.666	.023	-.947	.025	-.699	.022	-.710	.018	-.536
-.567	-.216	.035	-.638	.068	-.635	.079	-.547	.075	-.526	.077	-.601
-.452	-.301	.105	-.499	.134	-.503	.133	-.444	.129	-.400	.129	-.331
-.311	-.331	.178	-.459	.209	-.414	.214	-.365	.201	-.351	.209	-.270
-.023	-.291	.286	-.383	.294	-.344	.295	-.312	.294	-.278	.293	-.227
.133	-.226	.396	-.306	.404	-.297	.407	-.284	.397	-.253	.494	-.195
.272	-.168	.514	-.248	.497	-.278	.502	-.274	.495	-.253	.590	-.204
.416	-.149	.618	-.229	.599	-.253	.601	-.264	.594	-.241	.693	-.219
.565	-.113	.733	-.190	.700	-.254	.698	-.273	.693	-.258	.777	-.225
.713	-.100	.835	-.165	.864	-.218	.863	-.250	.784	-.268	.861	-.204
.854	-.107	.919	-.120	.926	-.158	.923	-.191	.856	-.258	.918	-.183
.980	-.098	.987	-.037	.975	-.069	.977	-.063	.926	-.193	.972	-.072
1.074	-.085							.977	-.055		
1.122	-.043										
					LOWER SURFACE						
-.660	.089	-.022	.316	.024	.280	.025	.233	.019	.211	.020	.189
-.616	.065	.038	.102	.075	.060	.130	-.008	.066	.002	.076	-.055
-.572	.041	.101	.041	.297	-.064	.298	-.048	.136	-.029	.136	-.104
-.462	-.002	.185	-.025	.400	-.049	.397	-.050	.214	-.042	.221	-.087
-.329	-.011	.398	-.043	.604	-.053	.501	-.045	.292	-.044	.295	-.086
-.172	-.005	.737	.114	.785	.162	.603	.009	.403	-.037	.396	-.055
-.030	-.062			.967	.168	.703	.068	.489	-.064	.497	-.058
.128	-.103			1.000	.001	.784	.144	.594	-.045	.597	-.028
.418	-.030					.868	.228	.700	.071	.702	.077
.564	.019					.923	.269	.786	.168	.786	.158
.710	.110					.972	.168	.858	.242	.864	.221
.976	.237							.919	.258	.912	.219
1.072	.204							.967	.140		
1.110	.147										
CN=	.4027		.3816		.3762		.3735		.3406		.2805
CM=	.0143		-.0413		-.0588		-.0758		-.0750		-.0684

$\alpha = 4.00^{\circ}$; $C_L = 0.396$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.096	-.021	-.799	.023	-1.072	.025	-.855	.022	-.804	.018	-.708
-.567	-.256	.035	-.653	.068	-.702	.079	-.630	.075	-.586	.077	-.630
-.452	-.332	.105	-.556	.134	-.526	.133	-.514	.129	-.450	.129	-.378
-.311	-.347	.178	-.486	.209	-.449	.214	-.396	.201	-.359	.209	-.283
-.023	-.297	.286	-.393	.294	-.382	.295	-.338	.294	-.311	.293	-.249
.133	-.237	.396	-.331	.404	-.317	.407	-.304	.397	-.272	.494	-.204
.272	-.194	.514	-.258	.497	-.292	.502	-.289	.495	-.263	.590	-.214
.416	-.162	.618	-.236	.599	-.265	.601	-.270	.594	-.263	.693	-.223
.565	-.118	.733	-.194	.700	-.265	.698	-.286	.693	-.264	.777	-.238
.713	-.104	.835	-.163	.864	-.220	.863	-.256	.784	-.272	.861	-.209
.854	-.115	.919	-.119	.926	-.162	.923	-.188	.856	-.263	.918	-.182
.980	-.106	.987	-.041	.975	-.065	.977	-.064	.926	-.191	.972	-.073
1.074	-.085							.977	-.053		
1.122	-.048										
					LOWER SURFACE						
-.660	.093	-.022	.339	.024	.320	.025	.283	.019	.306	.020	.279
-.616	.072	.038	.145	.075	.107	.130	.030	.066	.055	.076	.009
-.572	.054	.101	.042	.297	-.043	.298	-.019	.136	-.004	.136	-.057
-.462	.019	.185	.000	.400	-.033	.397	-.032	.214	-.007	.221	-.066
-.329	.006	.398	-.025	.604	-.047	.501	-.037	.292	-.025	.295	-.061
-.172	-.001	.737	.112	.785	.164	.603	.025	.403	-.032	.396	-.044
-.030	-.056			.967	.161	.703	.075	.489	-.045	.497	-.048
.128	-.084			1.000	-.004	.784	.153	.594	-.045	.597	-.020
.418	-.018					.868	.231	.700	.081	.702	.079
.564	.026					.923	.270	.786	.171	.786	.154
.710	.111					.972	.166	.858	.247	.864	.217
.976	.238							.919	.256	.912	.222
1.072	.196							.967	.140		
1.110	.144										
CN=	.4399		.4148		.4151		.4208		.3788		.3201
CM=	.0271		-.0391		-.0577		-.0752		-.0744		-.0663

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(b) M = 0.50. Continued.

$\alpha = 4.97^o$; $C_L = 0.481$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.130	-.021	-1.107	.023	-1.258	.025	-1.131	.022	-1.098	.018	-.974
-.567	-.299	.035	-.804	.068	-.910	.079	-.785	.075	-.752	.077	-.756
-.452	-.367	.105	-.644	.134	-.631	.133	-.616	.129	-.561	.129	-.460
-.311	-.383	.178	-.550	.209	-.495	.214	-.468	.201	-.440	.209	-.345
-.023	-.322	.286	-.449	.294	-.432	.295	-.400	.294	-.360	.293	-.289
.133	-.253	.396	-.359	.404	-.357	.407	-.339	.397	-.310	.494	-.224
.272	-.224	.514	-.286	.497	-.321	.502	-.311	.495	-.296	.590	-.229
.416	-.169	.618	-.253	.599	-.297	.601	-.297	.594	-.278	.693	-.236
.565	-.143	.733	-.204	.700	-.276	.698	-.295	.693	-.280	.777	-.248
.713	-.121	.835	-.167	.864	-.223	.863	-.254	.784	-.277	.861	-.217
.854	-.121	.919	-.114	.926	-.145	.923	-.176	.856	-.256	.918	-.188
.980	-.115	.987	-.037	.975	-.065	.977	-.071	.926	-.185	.972	-.072
1.074	-.097							.977	-.054		
1.122	-.057										
					LOWER SURFACE						
-.660	.091	-.022	.373	.024	.393	.025	.365	.019	.373	.020	.388
-.616	.094	.038	.204	.075	.186	.130	.097	.066	.137	.076	.081
-.572	.084	.101	.115	.297	.017	.298	.031	.136	.059	.136	.004
-.462	.035	.185	.060	.400	-.003	.357	-.001	.214	.032	.221	-.024
-.329	.029	.398	.001	.604	-.033	.501	-.010	.292	.023	.295	-.027
-.172	.025	.737	.120	.765	.172	.603	.041	.403	.010	.396	-.019
-.030	-.034			.967	.157	.703	.086	.489	-.024	.497	-.032
.128	-.069			1.000	-.012	.784	.160	.594	-.017	.597	-.010
.418	.007					.868	.238	.700	.087	.702	.079
.564	.050					.923	.283	.786	.175	.786	.157
.710	.141					.972	.170	.858	.247	.864	.211
.976	.249							.919	.256	.912	.216
1.072	.202							.967	.138		
1.110	.156										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.5209	.4952	.4946	.5041	.4596	.3879
CM=	.0404	-.0342	-.0555	-.0732	-.0713	-.0625

$\alpha = 6.04^o$; $C_L = 0.572$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.191	-.021	-1.418	.023	-1.830	.025	-1.592	.022	-1.549	.018	-1.298
-.567	-.321	.035	-.956	.068	-1.077	.079	-.925	.075	-.909	.077	-.947
-.452	-.422	.105	-.725	.134	-.738	.133	-.715	.129	-.670	.129	-.563
-.311	-.415	.178	-.623	.209	-.581	.214	-.537	.201	-.515	.209	-.417
-.023	-.356	.286	-.506	.294	-.480	.295	-.440	.294	-.413	.293	-.336
.133	-.271	.396	-.407	.404	-.388	.407	-.386	.397	-.348	.494	-.253
.272	-.229	.514	-.310	.497	-.342	.502	-.340	.495	-.328	.590	-.245
.416	-.185	.618	-.274	.599	-.308	.601	-.312	.594	-.309	.693	-.248
.565	-.151	.733	-.208	.700	-.286	.698	-.302	.693	-.297	.777	-.255
.713	-.130	.835	-.165	.864	-.205	.863	-.250	.784	-.289	.861	-.221
.854	-.153	.919	-.104	.926	-.138	.923	-.169	.856	-.246	.918	-.185
.980	-.126	.987	-.039	.975	-.060	.977	-.065	.926	-.164	.972	-.074
1.074	-.098							.977	-.050		
1.122	-.062										
					LOWER SURFACE						
-.660	.092	-.022	.386	.024	.471	.025	.448	.019	.466	.020	.437
-.616	.117	.038	.250	.075	.267	.130	.144	.066	.205	.076	.169
-.572	.107	.101	.171	.297	.052	.298	.066	.136	.132	.136	.084
-.462	.071	.185	.107	.400	.023	.357	.034	.214	.084	.221	.030
-.329	.055	.398	.033	.604	-.016	.501	.017	.292	.053	.295	.005
-.172	.044	.737	.140	.765	.183	.603	.061	.403	.035	.396	-.004
-.030	-.003			.967	.159	.703	.099	.489	.003	.497	-.022
.128	-.034			1.000	-.023	.784	.160	.594	-.004	.597	-.000
.418	.030					.868	.244	.700	.097	.702	.082
.564	.062					.923	.289	.786	.178	.786	.151
.710	.147					.972	.167	.858	.256	.864	.207
.976	.260							.919	.266	.912	.209
1.072	.226							.967	.133		
1.110	.153										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.5943	.5668	.5839	.5886	.5477	.4637
CM=	.0578	-.0328	-.0481	-.0682	-.0664	-.0568

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Continued.

$\alpha = 6.99^{\circ}$; $C_L = 0.649$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.231	-.021	-1.707	.023	-2.349	.025	-2.028	.022	-2.008	.018	-1.694
-.567	-.375	.035	-1.125	.068	-1.240	.079	-1.065	.075	-1.039	.077	-1.072
-.452	-.423	.105	-.807	.134	-.823	.133	-.784	.129	-.754	.129	-.656
-.311	-.445	.178	-.677	.209	-.640	.214	-.600	.201	-.593	.209	-.464
-.023	-.372	.286	-.539	.294	-.517	.295	-.492	.294	-.456	.293	-.384
.133	-.295	.396	-.432	.404	-.423	.407	-.412	.397	-.386	.494	-.276
.272	-.254	.514	-.336	.497	-.372	.502	-.367	.495	-.351	.590	-.260
.416	-.214	.618	-.286	.599	-.326	.601	-.331	.594	-.326	.693	-.258
.565	-.169	.733	-.225	.700	-.295	.698	-.320	.693	-.302	.777	-.259
.713	-.148	.835	-.167	.864	-.202	.863	-.237	.784	-.282	.861	-.216
.854	-.149	.919	-.101	.926	-.130	.923	-.160	.856	-.242	.918	-.174
.980	-.139	.987	-.045	.975	-.070	.977	-.062	.926	-.149	.972	-.072
1.074	-.109							.977	-.045		
1.122	-.058										
					LOWER SURFACE						
-.660	.088	-.022	.378	.024	.504	.025	.507	.019	.498	.020	.490
-.616	.136	.038	.301	.075	.302	.130	.226	.066	.293	.076	.220
-.572	.140	.101	.208	.297	.096	.298	.084	.136	.182	.136	.112
-.462	.102	.185	.153	.400	.060	.397	.071	.214	.124	.221	.053
-.329	.086	.398	.057	.604	.004	.501	.042	.292	.092	.295	.023
-.172	.068	.737	.147	.785	.187	.603	.077	.403	.054	.396	.006
-.030	.027			.967	.153	.703	.107	.489	.025	.497	-.009
.128	-.006			1.000	-.037	.784	.164	.594	.007	.597	.004
.418	.057					.868	.252	.700	.098	.702	.081
.564	.090					.923	.292	.786	.181	.786	.148
.710	.167					.972	.166	.858	.254	.864	.205
.976	.271							.919	.262	.912	.212
1.072	.231							.967	.129		
1.110	.161										
CN=	.6758		.6587		.6631		.6687		.6211		.5247
CM=	.0689		-.0294		-.0437		-.0632		-.0594		-.0510

$\alpha = 8.00^{\circ}$; $C_L = 0.730$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.263	-.021	-2.287	.023	-2.607	.025	-2.205	.022	-1.602	.018	-1.445
-.567	-.426	.035	-1.244	.068	-1.363	.079	-1.221	.075	-1.398	.077	-.990
-.452	-.477	.105	-.890	.134	-1.094	.133	-.866	.129	-1.145	.129	-.841
-.311	-.465	.178	-.746	.209	-.693	.214	-.650	.201	-.727	.209	-.672
-.023	-.394	.286	-.587	.294	-.558	.295	-.529	.294	-.647	.293	-.477
.133	-.319	.396	-.466	.404	-.448	.407	-.434	.397	-.427	.494	-.309
.272	-.276	.514	-.364	.497	-.393	.502	-.377	.495	-.364	.590	-.262
.416	-.218	.618	-.256	.599	-.336	.601	-.338	.594	-.314	.693	-.239
.565	-.186	.733	-.232	.700	-.291	.698	-.312	.693	-.281	.777	-.214
.713	-.161	.835	-.165	.864	-.194	.863	-.210	.784	-.262	.861	-.175
.854	-.170	.919	-.107	.926	-.124	.923	-.150	.856	-.230	.918	-.158
.980	-.144	.987	-.056	.975	-.065	.977	-.067	.926	-.153	.972	-.088
1.074	-.115							.977	-.067		
1.122	-.068										
					LOWER SURFACE						
-.660	.085	-.022	.345	.024	.531	.025	.534	.019	.520	.020	.503
-.616	.152	.038	.337	.075	.367	.130	.269	.066	.329	.076	.279
-.572	.156	.101	.259	.297	.126	.298	.125	.136	.225	.136	.168
-.462	.130	.185	.180	.400	.083	.397	.085	.214	.161	.221	.088
-.329	.110	.398	.090	.604	.015	.501	.058	.292	.112	.295	.059
-.172	.093	.737	.155	.785	.191	.603	.094	.403	.079	.396	.029
-.030	.043			.967	.153	.703	.116	.489	.038	.497	.006
.128	.010			1.000	-.046	.784	.173	.594	.023	.597	.012
.418	.073					.868	.258	.700	.104	.702	.079
.564	.104					.923	.287	.786	.188	.786	.142
.710	.185					.972	.166	.858	.250	.864	.197
.976	.283							.919	.260	.912	.197
1.072	.233							.967	.121		
1.110	.163										
CN=	.7492		.7432		.7350		.7190		.6925		.5634
CM=	.0859		-.0242		-.0376		-.0584		-.0572		-.0478

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(b) M = 0.50. Continued.

$\alpha = 9.02^o$; $C_L = 0.809$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.340	-.021	-2.907	.023	-1.800	.025	-2.425	.022	-1.104	.018	-.628
-.567	-.470	.035	-1.383	.068	-1.639	.079	-1.795	.075	-1.040	.077	-.551
-.452	-.526	.105	-1.014	.134	-1.626	.133	-1.108	.129	-.988	.129	-.594
-.311	-.520	.178	-.831	.209	-1.375	.214	-.626	.201	-1.000	.209	-.550
-.023	-.430	.286	-.660	.294	-.677	.295	-.491	.294	-.938	.293	-.502
.133	-.343	.396	-.515	.404	-.502	.407	-.389	.397	-.839	.494	-.420
.272	-.298	.514	-.392	.497	-.366	.502	-.329	.495	-.701	.590	-.361
.416	-.257	.618	-.331	.599	-.321	.601	-.288	.594	-.507	.693	-.310
.565	-.208	.733	-.248	.700	-.261	.698	-.252	.693	-.443	.777	-.269
.713	-.188	.835	-.185	.864	-.190	.863	-.180	.784	-.334	.861	-.223
.854	-.165	.919	-.122	.926	-.140	.923	-.136	.856	-.245	.918	-.203
.980	-.165	.987	-.077	.975	-.072	.977	-.117	.926	-.222	.972	-.196
1.074	-.129							.977	-.148		
1.122	-.077										
				LOWER SURFACE							
-.660	.073	-.022	.320	.024	.536	.025	.545	.019	.504	.020	.509
-.616	.164	.038	.371	.075	.357	.130	.288	.066	.348	.076	.306
-.572	.176	.101	.301	.297	.149	.298	.142	.136	.232	.136	.200
-.462	.154	.185	.226	.400	.104	.397	.114	.214	.179	.221	.107
-.329	.137	.398	.110	.604	.028	.501	.068	.292	.139	.295	.068
-.172	.118	.737	.164	.785	.188	.603	.100	.403	.094	.396	.033
-.030	.077			.967	.148	.703	.112	.489	.053	.497	.014
.128	.040			1.000	-.064	.784	.162	.594	.037	.597	.012
.418	.094					.868	.250	.700	.112	.702	.075
.564	.124					.923	.290	.786	.182	.786	.140
.710	.205					.972	.143	.858	.243	.864	.199
.976	.295							.919	.242	.912	.197
1.072	.238							.967	.083		
1.110	.172										
CN=	.8435		.8469		.8267		.7560		.8203		.5340
CM=	.1003		-.0212		-.0358		-.0394		-.1031		-.0839

$\alpha = 10.05^o$; $C_L = 0.884$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.401	-.021	-3.075	.023	-1.664	.025	-2.465	.022	-1.009	.018	-.533
-.567	-.505	.035	-1.792	.068	-1.625	.079	-2.134	.075	-.979	.077	-.518
-.452	-.569	.105	-1.263	.134	-1.550	.133	-1.405	.129	-.956	.129	-.513
-.311	-.556	.178	-.919	.209	-1.544	.214	-.480	.201	-.962	.209	-.484
-.023	-.450	.286	-.712	.294	-1.282	.295	-.534	.294	-.955	.293	-.487
.133	-.372	.396	-.541	.404	-.726	.407	-.459	.397	-.879	.494	-.424
.272	-.321	.514	-.405	.497	-.408	.502	-.415	.495	-.814	.590	-.382
.416	-.275	.618	-.317	.599	-.354	.601	-.348	.594	-.727	.693	-.327
.565	-.234	.733	-.244	.700	-.297	.698	-.248	.693	-.604	.777	-.299
.713	-.203	.835	-.163	.864	-.193	.863	-.198	.784	-.501	.861	-.255
.854	-.205	.919	-.129	.926	-.138	.923	-.169	.856	-.340	.918	-.262
.980	-.174	.987	-.093	.975	-.103	.977	-.159	.926	-.247	.972	-.241
1.074	-.132							.977	-.214		
1.122	-.078										
				LOWER SURFACE							
-.660	.066	-.022	.237	.024	.549	.025	.543	.019	.511	.020	.520
-.616	.170	.038	.414	.075	.430	.130	.305	.066	.384	.076	.343
-.572	.196	.101	.341	.297	.176	.298	.159	.136	.270	.136	.214
-.462	.171	.185	.270	.400	.128	.397	.110	.214	.195	.221	.120
-.329	.161	.398	.150	.604	.040	.501	.069	.292	.144	.295	.080
-.172	.134	.737	.179	.785	.199	.603	.090	.403	.104	.396	.046
-.030	.097			.967	.146	.703	.092	.489	.061	.497	.013
.128	.068			1.000	-.035	.784	.141	.594	.034	.597	.007
.418	.132					.868	.229	.700	.108	.702	.074
.564	.147					.923	.262	.786	.173	.786	.130
.710	.219					.972	.088	.858	.242	.864	.184
.976	.306							.919	.237	.912	.188
1.072	.241							.967	.066		
1.110	.189										
CN=	.9245		.9420		.9407		.8099		.8923		.5331
CM=	.1122		-.0147		-.0507		-.0370		-.1355		-.0919

 [redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50. Concluded.

$\alpha = 11.07^{\circ}$; $C_L = 0.952$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.493	-.021	-2.937	.023	-1.029	.025	-2.363	.022	-.992	.018	-.509
-.567	-.573	.035	-2.425	.068	-1.494	.079	-2.075	.075	-1.038	.077	-.503
-.452	-.620	.105	-1.606	.134	-.945	.133	-1.015	.129	-.970	.129	-.471
-.311	-.587	.178	-.791	.209	-.932	.214	-.428	.201	-1.020	.209	-.456
-.023	-.474	.286	-.639	.294	-.850	.295	-.384	.294	-.999	.293	-.425
.133	-.393	.396	-.518	.404	-.812	.407	-.328	.397	-1.004	.494	-.365
.272	-.349	.514	-.397	.497	-.704	.502	-.288	.495	-.970	.590	-.338
.416	-.297	.618	-.304	.599	-.694	.601	-.311	.594	-.842	.693	-.327
.565	-.290	.733	-.225	.700	-.637	.698	-.253	.693	-.746	.777	-.306
.713	-.232	.835	-.158	.864	-.460	.863	-.258	.784	-.598	.861	-.284
.854	-.217	.919	-.106	.926	-.191	.923	-.201	.856	-.375	.918	-.284
.980	-.192	.987	-.078	.975	-.196	.977	-.233	.926	-.330	.972	-.279
1.074	-.137							.977	-.177		
1.122	-.080										
				LOWER SURFACE							
-.660	.044	-.022	.193	.024	.566	.025	.528	.019	.498	.020	.512
-.615	.184	.038	.446	.075	.459	.130	.306	.066	.389	.076	.352
-.572	.214	.101	.382	.297	.186	.298	.158	.136	.284	.136	.234
-.462	.209	.185	.305	.400	.139	.397	.104	.214	.203	.221	.139
-.329	.173	.398	.168	.604	.039	.501	.060	.292	.156	.295	.091
-.172	.166	.737	.184	.765	.169	.603	.073	.403	.108	.396	.055
-.030	.118			.967	.000	.703	.065	.489	.056	.497	.011
.126	.085			1.000	-.407	.784	.116	.594	.040	.597	.010
.418	.140					.868	.215	.700	.100	.702	.071
.564	.182					.923	.247	.786	.173	.786	.125
.710	.241					.972	.058	.858	.236	.864	.178
.976	.323							.919	.231	.912	.180
1.072	.253							.967	.062		
1.110	.193										
CN=	1.0112		.9874		.9111		.7276		.9722		.5131
CM=	.1263		-.0018		-.1126		-.0379		-.1574		-.0915

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80.

$\alpha = -4.96^{\circ}$; $C_L = -0.455$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	.022	-.021	.352	.023	.384	.025	.377	.022	.452	.018	.464
-.567	-.005	.035	.195	.068	.248	.079	.228	.075	.249	.077	.159
-.452	-.091	.105	.104	.134	.145	.133	.152	.129	.185	.129	.149
-.311	-.120	.178	.051	.209	.074	.214	.099	.201	.119	.209	.082
-.023	-.066	.286	.010	.294	.051	.295	.060	.294	.080	.293	.023
.133	-.014	.396	-.005	.404	.008	.407	.013	.397	.027	.494	-.087
.272	.004	.514	-.017	.497	-.028	.502	-.020	.495	-.030	.590	-.144
.416	.038	.618	-.046	.599	-.059	.601	-.068	.594	-.078	.693	-.207
.565	.045	.733	-.054	.700	-.107	.698	-.127	.693	-.147	.777	-.262
.713	.040	.835	-.081	.864	-.172	.863	-.235	.784	-.195	.861	-.284
.854	.004	.919	-.086	.926	-.139	.923	-.218	.856	-.233	.918	-.319
.980	-.019	.987	-.027	.975	-.039	.977	-.095	.926	-.242	.972	-.231
1.074	-.043							.977	-.169		
1.122	-.024										
				LOWER SURFACE							
-.660	-.089	-.022	-1.007	.024	-1.526	.025	-1.563	.019	-.785	.020	-.446
-.616	-.194	.038	-.839	.075	-1.439	.130	-1.442	.066	-.764	.076	-.366
-.572	-.226	.101	-.844	.297	-.606	.298	-.329	.136	-.755	.136	-.358
-.462	-.244	.185	-.679	.400	-.341	.397	-.250	.214	-.743	.221	-.329
-.329	-.212	.398	-.367	.604	-.222	.501	-.189	.292	-.713	.295	-.317
-.172	-.173	.737	-.009	.785	.077	.603	-.066	.403	-.678	.396	-.268
-.030	-.281			.967	.131	.703	.006	.489	-.673	.497	-.222
.128	-.366			1.000	.055	.784	.027	.594	-.650	.597	-.214
.418	-.318					.868	.073	.700	-.522	.702	-.225
.564	-.222					.923	.076	.786	-.424	.786	-.211
.710	-.069					.972	.072	.858	-.256	.864	-.197
.976	.195							.919	-.217	.912	-.202
1.072	.201							.967	-.072		
1.110	.158										
CN=	-.2562		-.4177		-.4334		-.3859		-.5630		-.1990
CM=	-.0815		-.0485		-.0756		-.0933		.0369		-.0220

$\alpha = -3.98^{\circ}$; $C_L = -0.372$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	.028	-.021	.310	.023	.335	.025	.326	.022	.420	.018	.460
-.567	-.002	.035	.135	.068	.182	.079	.190	.075	.209	.077	.129
-.452	-.091	.105	.067	.134	.089	.133	.122	.129	.142	.129	.132
-.311	-.139	.178	-.012	.209	.039	.214	.076	.201	.091	.209	.068
-.023	-.113	.286	-.031	.294	.010	.295	.029	.294	.052	.293	.012
.133	-.049	.396	-.043	.404	-.019	.407	-.012	.397	.011	.494	-.080
.272	-.015	.514	-.041	.497	-.045	.502	-.042	.495	-.038	.590	-.126
.416	.004	.618	-.062	.599	-.074	.601	-.086	.594	-.078	.693	-.185
.565	.031	.733	-.079	.700	-.116	.698	-.142	.693	-.142	.777	-.243
.713	.024	.835	-.097	.864	-.169	.863	-.214	.784	-.188	.861	-.243
.854	-.008	.919	-.091	.926	-.133	.923	-.187	.856	-.213	.918	-.257
.980	-.028	.987	-.024	.975	-.047	.977	-.051	.926	-.177	.972	-.151
1.074	-.049							.977	-.045		
1.122	-.023										
				LOWER SURFACE							
-.660	-.049	-.022	-.689	.024	-1.448	.025	-1.540	.019	-.801	.020	-.614
-.616	-.150	.038	-.798	.075	-1.286	.130	-1.340	.066	-.804	.076	-.440
-.572	-.208	.101	-.664	.297	-.404	.298	-.333	.136	-.768	.136	-.461
-.462	-.202	.185	-.582	.400	-.341	.397	-.286	.214	-.747	.221	-.341
-.329	-.189	.398	-.332	.604	-.207	.501	-.197	.292	-.715	.295	-.323
-.172	-.156	.737	.021	.785	.087	.603	-.041	.403	-.589	.396	-.291
-.030	-.253			.967	.169	.703	.044	.489	-.562	.497	-.251
.128	-.341			1.000	.047	.784	.057	.594	-.461	.597	-.227
.418	-.289					.868	.101	.700	-.298	.702	-.178
.564	-.191					.923	.125	.786	-.171	.786	-.149
.710	-.034					.972	.116	.858	-.056	.864	-.139
.976	.200							.919	.032	.912	-.162
1.072	.200							.967	.052		
1.110	.165										
CN=	-.1815		-.3125		-.3432		-.3461		-.4513		-.2205
CM=	-.0763		-.0508		-.0782		-.0959		-.0082		-.0236

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(c) M = 0.80. Continued.

$\alpha = -3.00^\circ$; $C_L = -0.278$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.035	-.021	.241	.023	.257	.025	.265	.022	.379	.018	.432
-.567	-.028	.035	.039	.068	.112	.079	.116	.075	.161	.077	.099
-.452	-.121	.105	-.010	.134	.027	.133	.065	.129	.106	.129	.106
-.311	-.174	.178	-.051	.209	-.021	.214	.019	.201	.046	.209	.056
-.023	-.142	.286	-.069	.294	-.031	.295	-.010	.294	.023	.293	-.004
.133	-.079	.396	-.076	.404	-.058	.407	-.041	.397	-.015	.494	-.081
.272	-.042	.514	-.071	.497	-.079	.502	-.072	.495	-.061	.590	-.114
.416	-.016	.618	-.093	.599	-.107	.601	-.108	.594	-.098	.693	-.153
.565	.005	.733	-.090	.700	-.135	.698	-.161	.693	-.156	.777	-.188
.713	.010	.835	-.112	.864	-.176	.863	-.216	.784	-.203	.861	-.178
.854	-.017	.919	-.104	.926	-.141	.923	-.169	.856	-.223	.918	-.167
.980	-.038	.987	-.027	.975	-.047	.977	-.039	.926	-.187	.972	-.053
1.074	-.060							.977	-.043		
1.122	-.017										
					LOWER SURFACE						
-.660	-.048	-.022	-.447	.024	-1.210	.025	-1.414	.019	-1.088	.020	-.916
-.616	-.128	.038	-.560	.075	-.904	.130	-.897	.066	-.953	.076	-.835
-.572	-.164	.101	-.593	.297	-.397	.298	-.364	.136	-.791	.136	-.712
-.462	-.183	.185	-.494	.400	-.330	.397	-.306	.214	-.633	.221	-.590
-.329	-.179	.398	-.311	.604	-.186	.501	-.228	.292	-.499	.295	-.512
-.172	-.143	.737	.030	.785	.078	.603	-.065	.403	-.403	.396	-.377
-.030	-.235			.967	.177	.703	.051	.489	-.270	.497	-.235
.128	-.318			1.000	.038	.784	.088	.594	-.185	.597	-.155
.418	-.248					.868	.139	.700	-.061	.702	-.053
.564	-.168					.923	.160	.786	.012	.786	-.040
.710	-.020					.972	.129	.858	.054	.864	.065
.976	.197							.919	.073	.912	.033
1.072	.199							.967	.086		
1.110	.154										
CN=	-.1096		-.2171		-.2398		-.2502		-.2964		-.2725
CM=	-.0645		-.0472		-.0708		-.0895		-.0605		-.0520

$\alpha = -2.02^\circ$; $C_L = -0.178$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.025	-.021	.183	.023	.144	.025	.168	.022	.303	.018	.387
-.567	-.045	.035	-.020	.068	.028	.079	.050	.075	.087	.077	.021
-.452	-.147	.105	-.066	.134	-.043	.133	-.012	.129	.047	.129	.060
-.311	-.186	.178	-.102	.209	-.070	.214	-.024	.201	.010	.209	.011
-.023	-.164	.286	-.119	.294	-.100	.295	-.049	.294	-.018	.293	-.039
.133	-.104	.396	-.122	.404	-.101	.407	-.075	.397	-.046	.494	-.091
.272	-.075	.514	-.103	.497	-.128	.502	-.109	.495	-.078	.590	-.128
.416	-.041	.618	-.120	.599	-.134	.601	-.135	.594	-.122	.693	-.155
.565	-.012	.733	-.114	.700	-.165	.698	-.183	.693	-.168	.777	-.181
.713	-.017	.835	-.127	.864	-.190	.863	-.229	.784	-.207	.861	-.165
.854	-.045	.919	-.105	.926	-.146	.923	-.174	.856	-.227	.918	-.148
.980	-.053	.987	-.036	.975	-.050	.977	-.033	.926	-.181	.972	-.029
1.074	-.062							.977	-.021		
1.122	-.027										
					LOWER SURFACE						
-.660	-.001	-.022	-.302	.024	-.809	.025	-1.179	.019	-1.217	.020	-1.280
-.616	-.096	.038	-.477	.075	-.759	.130	-.490	.066	-1.096	.076	-1.166
-.572	-.133	.101	-.476	.297	-.339	.298	-.343	.136	-.602	.136	-.630
-.462	-.156	.185	-.385	.400	-.297	.397	-.294	.214	-.433	.221	-.399
-.329	-.155	.398	-.274	.604	-.175	.501	-.217	.292	-.386	.295	-.290
-.172	-.127	.737	.053	.785	.103	.603	-.071	.403	-.291	.396	-.211
-.030	-.200			.967	.187	.703	.048	.489	-.232	.497	-.138
.128	-.275			1.000	.034	.784	.105	.594	-.162	.597	-.050
.418	-.205					.868	.167	.700	-.006	.702	.048
.564	-.134					.923	.183	.786	.095	.786	.097
.710	.004					.972	.144	.858	.158	.864	.126
.976	.206							.919	.180	.912	.144
1.072	.206							.967	.134		
1.110	.155										
CN=	-.0205		-.1174		-.1344		-.1314		-.1998		-.1770
CM=	-.0565		-.0511		-.0720		-.0845		-.0805		-.0854

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(c) M = 0.80. Continued.

$\alpha = -1.03^o$; $C_L = -0.075$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.023	-.021	.076	.023	.005	.025	.059	.022	.197	.018	.330
-.567	-.063	.035	-.126	.068	-.086	.079	-.028	.075	-.002	.077	-.071
-.452	-.191	.105	-.156	.134	-.142	.133	-.076	.129	-.013	.129	-.010
-.311	-.236	.178	-.189	.209	-.146	.214	-.091	.201	-.062	.209	-.040
-.023	-.207	.286	-.175	.294	-.141	.295	-.112	.294	-.065	.293	-.076
.133	-.132	.396	-.170	.404	-.133	.407	-.125	.397	-.085	.494	-.109
.272	-.098	.514	-.139	.497	-.155	.502	-.141	.495	-.119	.590	-.136
.416	-.071	.618	-.153	.599	-.170	.601	-.170	.594	-.152	.693	-.171
.565	-.048	.733	-.140	.700	-.192	.698	-.213	.693	-.194	.777	-.189
.713	-.034	.835	-.146	.864	-.212	.863	-.239	.784	-.231	.861	-.170
.854	-.056	.919	-.121	.926	-.154	.923	-.177	.856	-.247	.918	-.141
.980	-.066	.987	-.030	.975	-.059	.977	-.031	.926	-.183	.972	-.013
1.074	-.073							.977	-.025		
1.122	-.031										

LOWER SURFACE

X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.022	-.022	-.171	.024	-.504	.025	-.898	.019	-1.027	.020	-1.047
-.616	-.060	.038	-.331	.075	-.522	.130	-.440	.066	-.860	.076	-.879
-.572	-.114	.101	-.347	.297	-.285	.298	-.309	.136	-.430	.136	-.631
-.462	-.139	.185	-.322	.400	-.246	.397	-.259	.214	-.365	.221	-.346
-.329	-.129	.398	-.229	.604	-.153	.501	-.203	.292	-.295	.295	-.265
-.172	-.111	.737	.067	.785	.119	.603	-.062	.403	-.235	.396	-.200
-.030	-.166			.967	.191	.703	.049	.489	-.218	.497	-.142
.128	-.253			1.000	.023	.764	.112	.594	-.151	.597	-.046
.418	-.176					.868	.181	.700	.017	.702	.079
.564	-.106					.923	.196	.786	.114	.786	.136
.710	.030					.972	.142	.858	.170	.864	.169
.976	.214							.919	.201	.912	.164
1.072	.203							.967	.150		
1.110	.158										

	.133	.307	.480	.653	.804	.933
CN=	.0661	-.0108	-.0182	-.0480	-.0953	-.1043
CM=	-.0469	-.0521	-.0706	-.0846	-.0834	-.0855

$\alpha = 0.03^o$; $C_L = -0.034$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.004	-.021	-.010	.023	-.214	.025	-.097	.022	.080	.018	.210
-.567	-.096	.035	-.228	.068	-.232	.079	-.154	.075	-.104	.077	-.202
-.452	-.210	.105	-.240	.134	-.219	.133	-.183	.129	-.111	.129	-.090
-.311	-.268	.178	-.264	.209	-.215	.214	-.168	.201	-.130	.209	-.099
-.023	-.224	.286	-.256	.294	-.200	.295	-.175	.294	-.122	.293	-.111
.133	-.155	.396	-.214	.404	-.195	.407	-.165	.397	-.140	.494	-.141
.272	-.128	.514	-.176	.497	-.197	.502	-.182	.495	-.172	.590	-.163
.416	-.092	.618	-.175	.599	-.199	.601	-.207	.594	-.194	.693	-.190
.565	-.059	.733	-.163	.700	-.223	.698	-.242	.693	-.230	.777	-.211
.713	-.053	.835	-.161	.864	-.228	.863	-.249	.784	-.252	.861	-.183
.854	-.078	.919	-.121	.926	-.163	.923	-.180	.856	-.263	.918	-.163
.980	-.084	.987	-.033	.975	-.066	.977	-.032	.926	-.190	.972	-.028
1.074	-.074							.977	-.028		
1.122	-.031										

LOWER SURFACE

X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.033	-.022	-.018	.024	-.284	.025	-.404	.019	-.643	.020	-.576
-.616	-.040	.038	-.219	.075	-.355	.130	-.349	.066	-.563	.076	-.600
-.572	-.068	.101	-.260	.297	-.259	.298	-.232	.136	-.414	.136	-.526
-.462	-.098	.185	-.246	.400	-.208	.397	-.222	.214	-.285	.221	-.270
-.329	-.099	.398	-.188	.604	-.131	.501	-.179	.292	-.241	.295	-.218
-.172	-.090	.737	.085	.785	.138	.603	-.051	.403	-.200	.396	-.183
-.030	-.169			.967	.187	.703	.059	.489	-.192	.497	-.129
.128	-.223			1.000	.019	.764	.129	.594	-.150	.597	-.052
.418	-.147					.868	.178	.700	.031	.702	.085
.564	-.081					.923	.201	.786	.140	.786	.159
.710	.044					.972	.153	.858	.202	.864	.204
.976	.219							.919	.228	.912	.203
1.072	.209							.967	.159		
1.110	.157										

	.133	.307	.480	.653	.804	.933
CN=	.1452	.0516	.0873	.0738	.0146	.0057
CM=	-.0333	-.0534	-.0707	-.0800	-.0858	-.0851

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(c) M = 0.80. Continued.

$\alpha = 1.02^\circ$; $C_L = 0.130$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.006	-.021	-.154	.023	-.414	.025	-.250	.022	-.065	.018	.054
-.567	-.129	.035	-.326	.068	-.372	.079	-.300	.075	-.224	.077	-.311
-.452	-.240	.105	-.326	.134	-.301	.133	-.255	.129	-.196	.129	-.162
-.311	-.299	.178	-.325	.209	-.266	.214	-.226	.201	-.193	.209	-.141
-.023	-.256	.286	-.305	.294	-.256	.295	-.205	.294	-.175	.293	-.154
.133	-.181	.396	-.254	.404	-.231	.407	-.207	.397	-.176	.494	-.163
.272	-.143	.514	-.211	.497	-.223	.502	-.225	.495	-.203	.590	-.182
.416	-.118	.618	-.202	.595	-.227	.601	-.222	.594	-.218	.693	-.204
.565	-.084	.733	-.177	.700	-.242	.658	-.270	.693	-.252	.777	-.224
.713	-.063	.835	-.164	.864	-.229	.863	-.263	.784	-.275	.861	-.197
.854	-.091	.919	-.125	.926	-.158	.923	-.178	.856	-.273	.918	-.163
.980	-.095	.987	-.031	.975	-.061	.977	-.036	.926	-.187	.972	-.034
1.074	-.086							.977	-.038		
1.122	-.035										
				LOWER SURFACE							
-.660	.055	-.022	.111	.024	-.067	.025	-.220	.019	-.352	.020	-.304
-.616	-.011	.038	-.117	.075	-.256	.130	-.270	.066	-.425	.076	-.435
-.572	-.045	.101	-.181	.297	-.206	.298	-.190	.136	-.288	.136	-.384
-.462	-.081	.185	-.187	.400	-.156	.397	-.167	.214	-.237	.221	-.255
-.329	-.082	.398	-.156	.604	-.118	.501	-.145	.292	-.181	.295	-.186
-.172	-.050	.737	.102	.765	.153	.603	-.039	.403	-.166	.396	-.159
-.030	-.139			.967	.185	.703	.056	.489	-.166	.497	-.114
.128	-.198			1.000	.012	.784	.140	.594	-.124	.597	-.046
.418	-.107					.868	.203	.700	.046	.702	.087
.564	-.058					.923	.231	.786	.159	.786	.174
.710	.068					.972	.177	.858	.226	.864	.223
.976	.238							.919	.259	.912	.235
1.072	.218							.967	.157		
1.110	.161										
CN=	.2307		.1635		.1759		.1633		.1117		.0900
CM=	-.0242		-.0529		-.0688		-.0826		-.0872		-.0843

$\alpha = 2.01^\circ$; $C_L = 0.231$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.030	-.021	-.331	.023	-.632	.025	-.445	.022	-.350	.018	-.145
-.567	-.157	.035	-.451	.068	-.579	.079	-.403	.075	-.362	.077	-.452
-.452	-.277	.105	-.448	.134	-.408	.133	-.363	.129	-.318	.129	-.252
-.311	-.311	.178	-.388	.209	-.362	.214	-.296	.201	-.254	.209	-.216
-.023	-.290	.286	-.348	.294	-.313	.295	-.275	.294	-.241	.293	-.209
.133	-.203	.396	-.293	.404	-.287	.407	-.256	.397	-.210	.494	-.195
.272	-.178	.514	-.243	.497	-.257	.502	-.259	.495	-.237	.590	-.196
.416	-.146	.618	-.226	.599	-.256	.601	-.266	.594	-.245	.693	-.220
.565	-.100	.733	-.193	.700	-.262	.698	-.286	.693	-.275	.777	-.234
.713	-.089	.835	-.177	.864	-.227	.863	-.267	.784	-.286	.861	-.207
.854	-.108	.919	-.121	.926	-.152	.923	-.181	.856	-.279	.918	-.169
.980	-.109	.987	-.029	.975	-.063	.977	-.045	.926	-.189	.972	-.034
1.074	-.094							.977	-.036		
1.122	-.046										
				LOWER SURFACE							
-.660	.065	-.022	.205	.024	.089	.025	-.017	.019	-.057	.020	-.090
-.616	.016	.038	-.029	.075	-.109	.130	-.165	.066	-.239	.076	-.294
-.572	-.003	.101	-.096	.297	-.133	.298	-.138	.136	-.218	.136	-.281
-.462	-.051	.185	-.122	.400	-.127	.397	-.132	.214	-.165	.221	-.217
-.329	-.052	.398	-.117	.604	-.099	.501	-.124	.292	-.150	.295	-.172
-.172	-.038	.737	.105	.765	.159	.603	-.029	.403	-.108	.396	-.127
-.030	-.117			.967	.183	.703	.056	.489	-.133	.497	-.106
.128	-.174			1.000	-.004	.784	.144	.594	-.111	.597	-.039
.418	-.094					.868	.228	.700	.049	.702	.088
.564	-.031					.923	.265	.786	.168	.786	.180
.710	.083					.972	.177	.858	.244	.864	.236
.976	.244							.919	.261	.912	.241
1.072	.220							.967	.160		
1.110	.160										
CN=	.3087		.2715		.2807		.2619		.2153		.1739
CM=	-.0122		-.0499		-.0650		-.0824		-.0839		-.0812

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Continued.

$\alpha = 2.47^{\circ}$; $C_L = 0.276$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	-.027	-.021	-.388	.023	-.730	.025	-.615	.022	-.419	.018	-.256
-.567	-.158	.035	-.510	.068	-.731	.079	-.480	.075	-.434	.077	-.515
-.452	-.275	.105	-.461	.134	-.455	.133	-.395	.129	-.356	.129	-.303
-.311	-.342	.178	-.434	.209	-.376	.214	-.343	.201	-.300	.209	-.244
-.023	-.297	.286	-.368	.294	-.330	.295	-.297	.294	-.259	.293	-.221
.133	-.222	.396	-.320	.404	-.300	.407	-.275	.397	-.246	.494	-.198
.272	-.194	.514	-.251	.497	-.273	.502	-.280	.495	-.246	.590	-.202
.416	-.160	.618	-.237	.599	-.260	.601	-.278	.594	-.262	.693	-.225
.565	-.119	.733	-.197	.700	-.273	.698	-.299	.693	-.277	.777	-.242
.713	-.105	.835	-.178	.864	-.233	.863	-.276	.784	-.293	.861	-.211
.854	-.116	.919	-.122	.926	-.152	.923	-.187	.856	-.280	.918	-.176
.980	-.109	.987	-.026	.975	-.064	.977	-.047	.926	-.184	.972	-.040
1.074	-.093							.977	-.038		
1.122	-.042										
				LOWER SURFACE							
-.660	.083	-.022	.251	.024	.156	.025	.052	.019	-.039	.020	-.015
-.616	.031	.038	.015	.075	-.072	.130	-.118	.066	-.138	.076	-.227
-.572	.001	.101	-.064	.257	-.101	.298	-.115	.136	-.168	.136	-.226
-.462	-.035	.185	-.090	.400	-.105	.357	-.121	.214	-.135	.221	-.180
-.329	-.047	.398	-.085	.604	-.088	.501	-.103	.292	-.133	.295	-.164
-.172	-.036	.737	.116	.785	.163	.603	-.016	.403	-.103	.396	-.110
-.030	-.101			.967	.189	.703	.056	.489	-.117	.497	-.097
.123	-.146			1.000	-.002	.784	.142	.594	-.098	.597	-.033
.418	-.074					.868	.221	.700	.061	.702	.090
.564	-.015					.923	.267	.786	.172	.786	.177
.710	.093					.972	.179	.858	.243	.864	.230
.976	.252							.919	.266	.912	.234
1.072	.226							.967	.149		
1.110	.169										
CN=	.3535		.3182		.3254		.3104		.2559		.2103
CM=	-.0104		-.0511		-.0648		-.0814		-.0833		-.0783

$\alpha = 2.94^{\circ}$; $C_L = 0.321$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	-.036	-.021	-.499	.023	-.986	.025	-.691	.022	-.531	.018	-.374
-.567	-.182	.035	-.571	.068	-.777	.079	-.618	.075	-.577	.077	-.644
-.452	-.291	.105	-.496	.134	-.513	.133	-.459	.129	-.410	.129	-.332
-.311	-.340	.178	-.473	.209	-.410	.214	-.371	.201	-.346	.209	-.280
-.023	-.306	.286	-.394	.294	-.354	.295	-.324	.294	-.282	.293	-.244
.133	-.216	.396	-.334	.404	-.323	.407	-.288	.397	-.261	.494	-.210
.272	-.206	.514	-.264	.497	-.292	.502	-.282	.495	-.263	.590	-.220
.416	-.162	.618	-.247	.599	-.282	.601	-.288	.594	-.271	.693	-.237
.565	-.113	.733	-.203	.700	-.274	.698	-.298	.693	-.293	.777	-.244
.713	-.107	.835	-.177	.864	-.228	.863	-.275	.784	-.306	.861	-.211
.854	-.121	.919	-.119	.926	-.146	.923	-.185	.856	-.280	.918	-.172
.980	-.123	.987	-.026	.975	-.056	.977	-.055	.926	-.184	.972	-.038
1.074	-.100							.977	-.038		
1.122	-.047										
				LOWER SURFACE							
-.660	.082	-.022	.280	.024	.206	.025	.121	.019	.108	.020	.092
-.616	.047	.038	.047	.075	-.026	.130	-.080	.066	-.116	.076	-.169
-.572	.015	.101	-.017	.257	-.095	.298	-.101	.136	-.127	.136	-.191
-.462	-.018	.185	-.068	.400	-.103	.357	-.105	.214	-.102	.221	-.164
-.329	-.039	.398	-.086	.604	-.082	.501	-.094	.292	-.103	.295	-.139
-.172	-.023	.737	.116	.785	.165	.603	-.005	.403	-.079	.396	-.102
-.030	-.095			.967	.179	.703	.063	.489	-.102	.497	-.090
.128	-.141			1.000	-.012	.784	.150	.594	-.081	.597	-.040
.418	-.062					.868	.229	.700	.060	.702	.088
.564	-.009					.923	.270	.786	.173	.786	.182
.710	.094					.972	.177	.858	.250	.864	.228
.976	.258							.919	.269	.912	.236
1.072	.221							.967	.155		
1.110	.166										
CN=	.3768		.3516		.3644		.3515		.3073		.2515
CM=	-.0044		-.0479		-.0602		-.0796		-.0823		-.0752

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Continued.

$\alpha = 3.45^{\circ}$; $C_L = 0.370$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.061	-.021	-.637	.023	-1.142	.025	-.832	.022	-.674	.018	-.522
-.567	-.205	.035	-.626	.068	-1.117	.079	-.742	.075	-.639	.077	-.695
-.452	-.312	.105	-.596	.134	-.544	.133	-.511	.129	-.465	.129	-.403
-.311	-.361	.178	-.520	.209	-.450	.214	-.405	.201	-.382	.209	-.299
-.023	-.327	.286	-.438	.294	-.386	.295	-.362	.294	-.330	.293	-.264
.133	-.247	.396	-.358	.404	-.345	.407	-.313	.397	-.282	.494	-.218
.272	-.209	.514	-.289	.497	-.305	.502	-.305	.495	-.289	.590	-.225
.416	-.172	.618	-.249	.599	-.284	.601	-.302	.594	-.291	.693	-.238
.565	-.131	.733	-.215	.700	-.285	.698	-.310	.693	-.299	.777	-.251
.713	-.116	.835	-.180	.864	-.236	.863	-.273	.784	-.298	.861	-.219
.854	-.138	.919	-.117	.926	-.147	.923	-.179	.856	-.282	.918	-.178
.980	-.130	.987	-.032	.975	-.067	.977	-.055	.926	-.179	.972	-.038
1.074	-.108							.977	-.035		
1.122	-.055										
					LOWER SURFACE						
-.660	.094	-.022	.314	.024	.260	.025	.187	.019	.180	.020	.166
-.616	.053	.038	.098	.075	.025	.130	-.034	.066	-.039	.076	-.100
-.572	.030	.101	.005	.297	-.075	.298	-.076	.136	-.069	.136	-.159
-.462	-.024	.185	-.038	.400	-.075	.397	-.084	.214	-.085	.221	-.125
-.329	-.017	.398	-.057	.604	-.077	.501	-.072	.292	-.083	.295	-.129
-.172	-.014	.737	.119	.785	.167	.603	-.003	.403	-.063	.396	-.094
-.030	-.075			.967	.184	.703	.068	.489	-.087	.497	-.084
.128	-.133			1.000	-.011	.784	.156	.594	-.080	.597	-.033
.418	-.048					.868	.235	.700	.070	.702	.085
.564	.005					.923	.278	.786	.177	.786	.171
.710	.115					.972	.177	.858	.254	.864	.226
.976	.266							.919	.269	.912	.230
1.072	.225							.967	.147		
1.110	.174										
CN=	.4266		.4069		.4246		.4038		.3546		.2875
CM=	.0003		-.0465		-.0573		-.0785		-.0801		-.0715

$\alpha = 4.02^{\circ}$; $C_L = 0.427$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.061	-.021	-.774	.023	-1.257	.025	-1.167	.022	-.879	.018	-.707
-.567	-.215	.035	-.812	.068	-1.283	.079	-.828	.075	-.773	.077	-.795
-.452	-.338	.105	-.631	.134	-.601	.133	-.567	.129	-.520	.129	-.458
-.311	-.389	.178	-.553	.209	-.478	.214	-.444	.201	-.429	.209	-.351
-.023	-.335	.286	-.463	.294	-.413	.295	-.378	.294	-.352	.293	-.290
.133	-.245	.396	-.391	.404	-.353	.407	-.333	.397	-.312	.494	-.235
.272	-.232	.514	-.302	.497	-.326	.502	-.323	.495	-.312	.590	-.231
.416	-.198	.618	-.272	.599	-.292	.601	-.322	.594	-.312	.693	-.243
.565	-.155	.733	-.222	.700	-.291	.698	-.330	.693	-.315	.777	-.257
.713	-.131	.835	-.180	.864	-.235	.863	-.277	.784	-.310	.861	-.224
.854	-.137	.919	-.111	.926	-.152	.923	-.173	.856	-.283	.918	-.182
.980	-.135	.987	-.031	.975	-.065	.977	-.047	.926	-.176	.972	-.046
1.074	-.112							.977	-.039		
1.122	-.054										
					LOWER SURFACE						
-.660	.056	-.022	.342	.024	.306	.025	.275	.019	.250	.020	.242
-.616	.065	.038	.145	.075	.074	.130	.002	.066	.039	.076	-.028
-.572	.045	.101	.042	.297	-.037	.298	-.048	.136	-.038	.136	-.090
-.462	.001	.185	-.012	.400	-.054	.397	-.063	.214	-.047	.221	-.109
-.329	-.012	.398	-.046	.604	-.057	.501	-.065	.292	-.046	.295	-.103
-.172	-.009	.737	.125	.785	.169	.603	.008	.403	-.052	.396	-.074
-.030	-.066			.967	.185	.703	.070	.489	-.078	.497	-.076
.128	-.121			1.000	-.017	.784	.157	.594	-.071	.597	-.024
.418	-.034					.868	.239	.700	.074	.702	.090
.564	.021					.923	.290	.786	.182	.786	.172
.710	.125					.972	.188	.858	.254	.864	.231
.976	.271							.919	.273	.912	.227
1.072	.228							.967	.150		
1.110	.174										
CN=	.4732		.4551		.4752		.4635		.4114		.3394
CM=	.0060		-.0435		-.0562		-.0758		-.0787		-.0690

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Continued.

$\alpha = 4.97^{\circ}$; $C_L = 0.524$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.106	-.021	-1.141	.023	-1.441	.025	-1.397	.022	-1.230	.018	-1.111
-.567	-.245	.035	-1.013	.068	-1.572	.079	-1.430	.075	-1.175	.077	-1.042
-.452	-.361	.105	-.704	.134	-.891	.133	-.727	.129	-.628	.129	-.544
-.311	-.410	.178	-.647	.209	-.534	.214	-.480	.201	-.478	.209	-.392
-.023	-.363	.286	-.525	.294	-.441	.295	-.417	.294	-.398	.293	-.316
.133	-.289	.396	-.418	.404	-.362	.407	-.370	.397	-.338	.494	-.251
.272	-.239	.514	-.314	.497	-.341	.502	-.347	.495	-.334	.590	-.248
.416	-.215	.618	-.280	.599	-.314	.601	-.321	.594	-.320	.693	-.262
.565	-.172	.733	-.224	.700	-.296	.698	-.323	.693	-.320	.777	-.268
.713	-.143	.835	-.181	.864	-.230	.863	-.258	.784	-.305	.861	-.231
.854	-.162	.919	-.096	.926	-.141	.923	-.169	.856	-.274	.918	-.178
.980	-.143	.987	-.033	.975	-.060	.977	-.053	.926	-.166	.972	-.039
1.074	-.122							.977	-.040		
1.122	-.057										
					LOWER SURFACE						
-.660	.102	-.022	.383	.024	.378	.025	.339	.019	.334	.020	.328
-.616	.089	.038	.192	.075	.155	.130	.057	.066	.106	.076	.036
-.572	.069	.101	.102	.297	-.007	.298	-.010	.136	.037	.136	-.038
-.462	.021	.185	.034	.400	-.023	.397	-.037	.214	.002	.221	-.055
-.329	.013	.398	-.008	.604	-.051	.501	-.038	.292	-.014	.295	-.065
-.172	.019	.737	.138	.785	.187	.603	.030	.403	-.012	.396	-.062
-.030	-.045			.967	.191	.703	.083	.489	-.056	.497	-.062
.128	-.083			1.000	-.006	.784	.160	.594	-.050	.597	-.025
.418	-.005					.868	.242	.700	.081	.702	.083
.564	.038					.923	.296	.786	.188	.786	.168
.710	.143					.972	.183	.858	.260	.864	.216
.976	.281							.919	.272	.912	.219
1.072	.237							.967	.144		
1.110	.179										
CN=	.5521		.5490		.5600		.5557		.4991		.4108
CM=	.0167		-.0376		-.0505		-.0667		-.0708		-.0606

$\alpha = 6.06^{\circ}$; $C_L = 0.628$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.153	-.021	-1.477	.023	-1.566	.025	-1.514	.022	-1.137	.018	-1.151
-.567	-.279	.035	-1.252	.068	-1.670	.079	-1.561	.075	-1.053	.077	-.989
-.452	-.412	.105	-.889	.134	-1.017	.133	-1.159	.129	-.865	.129	-.761
-.311	-.431	.178	-.774	.209	-.898	.214	-.678	.201	-.675	.209	-.539
-.023	-.394	.286	-.596	.294	-.751	.295	-.445	.294	-.539	.293	-.389
.133	-.311	.396	-.460	.404	-.424	.407	-.368	.397	-.447	.494	-.268
.272	-.281	.514	-.334	.497	-.360	.502	-.336	.495	-.378	.590	-.248
.416	-.253	.618	-.292	.599	-.302	.601	-.293	.594	-.339	.693	-.252
.565	-.195	.733	-.225	.700	-.289	.698	-.285	.693	-.302	.777	-.242
.713	-.174	.835	-.163	.864	-.215	.863	-.207	.784	-.275	.861	-.192
.854	-.163	.919	-.086	.926	-.148	.923	-.149	.856	-.230	.918	-.150
.980	-.159	.987	-.033	.975	-.065	.977	-.084	.926	-.142	.972	-.065
1.074	-.131							.977	-.071		
1.122	-.069										
					LOWER SURFACE						
-.660	.109	-.022	.420	.024	.437	.025	.411	.019	.408	.020	.411
-.616	.113	.038	.263	.075	.203	.130	.128	.066	.187	.076	.124
-.572	.111	.101	.167	.297	.038	.298	.042	.136	.081	.136	.014
-.462	.057	.185	.095	.400	.002	.397	-.002	.214	.050	.221	-.024
-.329	.047	.398	.035	.604	-.032	.501	-.019	.292	.026	.295	-.037
-.172	.043	.737	.155	.785	.196	.603	.041	.403	.000	.396	-.045
-.030	-.015			.967	.196	.703	.076	.489	-.030	.497	-.053
.128	-.053			1.000	.008	.784	.150	.594	-.038	.597	-.018
.418	.016					.868	.240	.700	.090	.702	.083
.564	.063					.923	.291	.786	.189	.786	.157
.710	.166					.972	.161	.858	.260	.864	.211
.976	.299							.919	.271	.912	.210
1.072	.246							.967	.141		
1.110	.186										
CN=	.6488		.6608		.6709		.6283		.5673		.4621
CM=	.0286		-.0305		-.0492		-.0511		-.0700		-.0535

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(c) M = 0.80. Continued.

$\alpha = 7.08^o$; $C_L = 0.719$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.169	-.021	-1.607	.023	-1.403	.025	-1.540	.022	-1.039	.018	-.939
-.567	-.330	.035	-1.645	.068	-1.494	.079	-1.610	.075	-1.032	.077	-.839
-.452	-.446	.105	-1.103	.134	-1.179	.133	-1.063	.129	-.974	.129	-.769
-.311	-.485	.178	-.970	.209	-1.073	.214	-.735	.201	-.891	.209	-.651
-.023	-.431	.266	-.710	.294	-.949	.295	-.511	.294	-.793	.293	-.550
.133	-.346	.396	-.489	.404	-.810	.407	-.413	.397	-.676	.494	-.329
.272	-.318	.514	-.357	.497	-.566	.502	-.358	.495	-.525	.590	-.260
.416	-.293	.618	-.313	.599	-.530	.601	-.308	.594	-.403	.693	-.223
.565	-.239	.733	-.235	.700	-.368	.698	-.275	.693	-.340	.777	-.197
.713	-.200	.835	-.166	.864	-.239	.863	-.217	.784	-.285	.861	-.146
.854	-.194	.919	-.090	.926	-.161	.923	-.182	.856	-.225	.918	-.129
.980	-.181	.987	-.036	.975	-.124	.977	-.176	.926	-.166	.972	-.100
1.074	-.145							.977	-.095		
1.122	-.076										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.110	-.022	.438	.024	.475	.025	.434	.019	.443	.020	.430
-.616	.134	.038	.308	.075	.273	.130	.149	.066	.214	.076	.165
-.572	.125	.101	.208	.297	.074	.298	.050	.136	.120	.136	.059
-.462	.080	.185	.139	.400	.040	.397	.011	.214	.079	.221	.000
-.329	.066	.398	.062	.604	-.013	.501	-.010	.292	.057	.295	-.024
-.172	.071	.737	.162	.785	.202	.603	.027	.403	.014	.396	-.043
-.030	.012			.967	.185	.703	.060	.489	-.021	.497	-.051
.128	-.028			1.000	-.002	.784	.133	.594	-.028	.597	-.027
.418	.051					.868	.235	.700	.091	.702	.069
.564	.090					.923	.289	.786	.186	.786	.145
.710	.190					.972	.133	.858	.260	.864	.183
.976	.306							.919	.271	.912	.191
1.072	.256							.967	.135		
1.110	.189										
CN=	.7525		.7732		.8239		.6531		.6756		.4801
CM=	.0410		-.0243		-.0797		-.0543		-.0830		-.0509

$\alpha = 8.14^o$; $C_L = 0.792$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.211	-.021	-1.696	.023	-1.008	.025	-1.474	.022	-.963	.018	-.627
-.567	-.370	.035	-1.744	.068	-1.050	.079	-1.474	.075	-.953	.077	-.609
-.452	-.472	.105	-1.288	.134	-.990	.133	-1.151	.129	-.927	.129	-.571
-.311	-.510	.178	-1.350	.209	-.792	.214	-.588	.201	-.890	.209	-.520
-.023	-.448	.266	-.858	.294	-.871	.295	-.467	.294	-.853	.293	-.506
.133	-.382	.396	-.444	.404	-.870	.407	-.396	.397	-.786	.494	-.395
.272	-.341	.514	-.330	.497	-.753	.502	-.353	.495	-.693	.590	-.356
.416	-.305	.618	-.305	.599	-.689	.601	-.320	.594	-.586	.693	-.306
.565	-.250	.733	-.225	.700	-.592	.698	-.293	.693	-.485	.777	-.276
.713	-.236	.835	-.160	.864	-.408	.863	-.259	.784	-.396	.861	-.231
.854	-.232	.919	-.088	.926	-.390	.923	-.268	.856	-.340	.918	-.203
.980	-.191	.987	-.035	.975	-.274	.977	-.242	.926	-.243	.972	-.198
1.074	-.160							.977	-.201		
1.122	-.092										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.116	-.022	.426	.024	.515	.025	.482	.019	.454	.020	.451
-.616	.152	.038	.351	.075	.317	.130	.180	.066	.257	.076	.183
-.572	.161	.101	.256	.297	.100	.298	.069	.136	.139	.136	.083
-.462	.115	.185	.189	.400	.056	.397	.017	.214	.089	.221	.020
-.329	.102	.398	.098	.604	-.013	.501	-.014	.292	.054	.295	-.018
-.172	.107	.737	.173	.785	.182	.603	.025	.403	.022	.396	-.040
-.030	.045			.967	.064	.703	.044	.489	-.012	.497	-.075
.128	.002			1.000	-.266	.784	.116	.594	-.030	.597	-.052
.418	.073					.868	.220	.700	.086	.702	.042
.564	.110					.923	.268	.786	.178	.786	.107
.710	.200					.972	.107	.858	.244	.864	.161
.976	.324							.919	.251	.912	.163
1.072	.271							.967	.100		
1.110	.192										
CN=	.8409		.8669		.8443		.6505		.7580		.4628
CM=	.0528		-.0175		-.1276		-.0604		-.1160		-.0735

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80. Continued.

$\alpha = 9.17^{\circ}$; $C_L = 0.857$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.247	-.021	-1.632	.023	-1.499	.025	-1.530	.022	-.944	.018	-.554
-.567	-.417	.035	-1.631	.068	-1.465	.079	-1.500	.075	-.983	.077	-.529
-.452	-.513	.105	-1.517	.134	-1.462	.133	-1.061	.129	-.957	.129	-.503
-.311	-.542	.178	-1.475	.209	-1.352	.214	-.465	.201	-.944	.209	-.472
-.023	-.487	.286	-1.265	.294	-1.242	.295	-.427	.294	-.918	.293	-.455
.133	-.407	.396	-.899	.404	-.556	.407	-.397	.397	-.865	.494	-.389
.272	-.361	.514	-.207	.497	-.679	.502	-.363	.495	-.767	.590	-.364
.416	-.322	.618	-.486	.599	-.630	.601	-.337	.594	-.738	.693	-.326
.565	-.278	.733	-.244	.700	-.620	.698	-.339	.693	-.575	.777	-.304
.713	-.240	.835	-.241	.864	-.549	.863	-.315	.784	-.563	.861	-.281
.854	-.244	.919	-.157	.926	-.533	.923	-.301	.856	-.437	.918	-.265
.980	-.217	.987	-.073	.975	-.482	.977	-.304	.926	-.405	.972	-.252
1.074	-.182							.977	-.335		
1.122	-.100										
				LOWER SURFACE							
-.660	.104	-.022	.431	.024	.536	.025	.504	.019	.505	.020	.457
-.616	.187	.038	.379	.075	.358	.130	.211	.066	.295	.076	.202
-.572	.167	.101	.300	.297	.116	.298	.082	.136	.187	.136	.095
-.462	.149	.165	.212	.400	.072	.397	.023	.214	.117	.221	.025
-.329	.130	.398	.115	.604	-.031	.501	.003	.292	.079	.295	-.013
-.172	.116	.737	.180	.785	.138	.603	.035	.403	.038	.396	-.046
-.030	.075			.967	.047	.703	.054	.489	-.008	.497	-.082
.128	.035			1.000	-.349	.784	.133	.594	-.027	.597	-.078
.418	.108					.868	.239	.700	.067	.702	.008
.564	.131					.923	.284	.786	.152	.786	.076
.710	.221					.972	.125	.858	.217	.864	.135
.976	.335							.919	.211	.912	.135
1.072	.267							.967	-.001		
1.110	.196										
CN=	.9299		1.0216		1.0348		.6714		.8482		.4419
CM=	.0671		-.0410		-.1282		-.0746		-.1411		-.0767

$\alpha = 10.22^{\circ}$; $C_L = 0.927$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.283	-.021	-1.349	.023	-1.559	.025	-1.564	.022	-.981	.018	-.507
-.567	-.460	.035	-1.349	.068	-1.553	.079	-1.532	.075	-.903	.077	-.483
-.452	-.589	.105	-1.229	.134	-1.519	.133	-1.425	.129	-.900	.129	-.475
-.311	-.566	.178	-1.170	.209	-1.482	.214	-.916	.201	-.897	.209	-.438
-.023	-.519	.286	-1.089	.294	-1.394	.295	-.555	.294	-.888	.293	-.436
.133	-.431	.396	-.970	.404	-.925	.407	-.362	.397	-.892	.494	-.396
.272	-.373	.514	-.855	.497	-.337	.502	-.377	.495	-.871	.590	-.362
.416	-.327	.618	-.745	.599	-.302	.601	-.333	.594	-.798	.693	-.335
.565	-.292	.733	-.338	.700	-.307	.698	-.308	.693	-.726	.777	-.332
.713	-.290	.835	-.470	.864	-.299	.863	-.299	.784	-.631	.861	-.315
.854	-.312	.919	-.339	.926	-.297	.923	-.289	.856	-.432	.918	-.309
.980	-.246	.987	-.247	.975	-.272	.977	-.296	.926	-.465	.972	-.295
1.074	-.202							.977	-.398		
1.122	-.140										
				LOWER SURFACE							
-.660	.122	-.022	.424	.024	.547	.025	.521	.019	.505	.020	.480
-.616	.210	.038	.409	.075	.376	.130	.237	.066	.324	.076	.237
-.572	.208	.101	.318	.297	.123	.298	.087	.136	.215	.136	.134
-.462	.169	.165	.258	.400	.067	.397	.048	.214	.137	.221	.045
-.329	.147	.398	.126	.604	-.043	.501	.005	.292	.107	.295	-.011
-.172	.158	.737	.171	.785	.120	.603	.044	.403	.055	.396	-.045
-.030	.101			.967	.003	.703	.058	.489	.008	.497	-.087
.128	.054			1.000	-.376	.784	.127	.594	-.023	.597	-.081
.418	.127					.868	.240	.700	.065	.702	-.005
.564	.149					.923	.289	.786	.143	.786	.061
.710	.240					.972	.127	.858	.203	.864	.126
.976	.343							.919	.190	.912	.118
1.072	.269							.967	-.032		
1.110	.185										
CN=	1.0237		1.0984		.9013		.7478		.8872		.4431
CM=	.0760		-.1150		-.0469		-.0664		-.1578		-.0805

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(c) M = 0.80. Concluded.

$\alpha = 11.22^o$; $C_L = 0.990$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.368	-.021	-1.215	.023	-1.381	.025	-1.529	.022	-.979	.018	-.473
-.567	-.503	.035	-1.167	.068	-1.066	.079	-1.527	.075	-.957	.077	-.460
-.452	-.599	.105	-1.183	.134	-1.346	.133	-1.558	.129	-.922	.129	-.466
-.311	-.620	.178	-1.142	.209	-1.273	.214	-.732	.201	-.949	.209	-.434
-.023	-.538	.286	-1.140	.294	-.995	.295	-.373	.294	-.940	.293	-.414
.133	-.460	.396	-.999	.404	-1.039	.407	-.369	.397	-.928	.494	-.383
.272	-.402	.514	-.926	.497	-.867	.502	-.354	.495	-.950	.590	-.366
.416	-.360	.618	-.791	.599	-.689	.601	-.326	.594	-.848	.693	-.346
.565	-.330	.733	-.690	.700	-.577	.698	-.329	.693	-.784	.777	-.353
.713	-.323	.835	-.546	.864	-.335	.863	-.314	.784	-.679	.861	-.345
.854	-.352	.919	-.425	.926	-.424	.923	-.307	.856	-.586	.918	-.334
.980	-.322	.987	-.299	.975	-.338	.977	-.309	.926	-.491	.972	-.329
1.074	-.241							.977	-.270		
1.122	-.147										
				LOWER SURFACE							
-.660	.101	-.022	.423	.024	.563	.025	.541	.019	.515	.020	.494
-.616	.211	.038	.462	.075	.412	.130	.266	.066	.353	.076	.271
-.572	.238	.101	.374	.297	.152	.298	.113	.136	.228	.136	.144
-.462	.218	.185	.278	.400	.102	.397	.058	.214	.153	.221	.059
-.329	.195	.398	.159	.604	-.030	.501	.005	.292	.112	.295	.015
-.172	.174	.737	.170	.785	.121	.603	.046	.403	.071	.396	-.041
-.030	.141			.967	-.009	.703	.046	.489	.009	.497	-.083
.128	.091			1.000	-.393	.784	.116	.594	-.022	.597	-.088
.418	.150					.868	.231	.700	.064	.702	-.002
.564	.177					.923	.275	.786	.138	.786	.064
.710	.249					.972	.090	.858	.200	.864	.115
.976	.345							.919	.191	.912	.118
1.072	.274							.967	-.025		
1.110	.183										
CN=	1.1329		1.1712		.9681		.7313		.9392		.4493
CM=	.0659		-.1531		-.1080		-.0647		-.1688		-.0849

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(d) M = 0.90.

$\alpha = -4.90^\circ$; $C_L = -0.482$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.029	-.021	.356	.023	.378	.025	.364	.022	.458	.018	.457
-.567	-.004	.035	.201	.068	.224	.079	.226	.075	.253	.077	.122
-.452	-.087	.105	.112	.134	.125	.133	.149	.129	.179	.129	.143
-.311	-.148	.178	.053	.209	.078	.214	.102	.201	.121	.209	.068
-.023	-.093	.286	.009	.294	.046	.295	.054	.294	.074	.293	-.003
.133	-.020	.396	-.014	.404	.012	.407	.013	.397	.024	.494	-.128
.272	.015	.514	-.001	.497	-.024	.502	-.032	.495	-.042	.590	-.199
.416	.039	.618	-.036	.599	-.058	.601	-.078	.594	-.099	.693	-.287
.565	.061	.733	-.051	.700	-.113	.698	-.162	.693	-.173	.777	-.390
.713	.062	.835	-.087	.864	-.207	.863	-.297	.784	-.259	.861	-.387
.854	.015	.919	-.074	.926	-.176	.923	-.288	.856	-.313	.918	-.324
.980	-.018	.987	.008	.975	-.087	.977	-.138	.926	-.350	.972	-.243
1.074	-.042							.977	-.361		
1.122	-.019										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.080	-.022	-.751	.024	-1.143	.025	-1.266	.019	-.744	.020	-.388
-.616	-.178	.038	-.996	.075	-1.169	.130	-1.272	.066	-.745	.076	-.365
-.572	-.220	.101	-.643	.297	-1.093	.298	-.741	.136	-.701	.136	-.348
-.462	-.226	.185	-.698	.400	-.586	.397	-.326	.214	-.708	.221	-.339
-.329	-.208	.398	-.689	.604	-.159	.501	-.216	.292	-.708	.295	-.319
-.172	-.137	.737	-.048	.785	.104	.603	-.151	.403	-.699	.396	-.283
-.030	-.253			.967	.143	.703	-.070	.489	-.694	.497	-.247
.128	-.371			1.000	.005	.784	-.009	.594	-.631	.597	-.240
.418	-.358					.868	.016	.700	-.561	.702	-.237
.564	-.363					.923	.010	.786	-.480	.786	-.243
.710	-.139					.972	.007	.858	-.413	.864	-.233
.976	.195							.919	-.329	.912	-.230
1.072	.222							.967	-.267		
1.110	.182										

	.133	.307	.480	.653	.804	.933
CN=	-.2868	-.5169	-.4679	-.4137	-.5577	-.1667
CM=	-.0559	-.0212	-.0691	-.0770	.0410	-.0312

$\alpha = -3.82^\circ$; $C_L = -0.389$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.022	-.021	.295	.023	.287	.025	.308	.022	.402	.018	.433
-.567	-.010	.035	.116	.068	.149	.079	.162	.075	.189	.077	.093
-.452	-.112	.105	.043	.134	.056	.133	.101	.129	.138	.129	.113
-.311	-.178	.178	-.014	.209	.025	.214	.059	.201	.084	.209	.060
-.023	-.132	.286	-.043	.294	-.004	.295	.021	.294	.042	.293	-.015
.133	-.058	.396	-.058	.404	-.038	.407	-.019	.397	-.001	.494	-.121
.272	-.021	.514	-.041	.497	-.064	.502	-.056	.495	-.054	.590	-.190
.416	.006	.618	-.074	.599	-.093	.601	-.100	.594	-.110	.693	-.272
.565	.045	.733	-.083	.700	-.138	.698	-.166	.693	-.180	.777	-.361
.713	.032	.835	-.111	.864	-.198	.863	-.249	.784	-.236	.861	-.309
.854	-.013	.919	-.099	.926	-.156	.923	-.209	.856	-.244	.918	-.332
.980	-.044	.987	-.011	.975	-.059	.977	-.069	.926	-.210	.972	-.217
1.074	-.058							.977	-.136		
1.122	-.026										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.056	-.022	-.623	.024	-1.062	.025	-1.190	.019	-.898	.020	-.483
-.616	-.141	.038	-.698	.075	-1.092	.130	-1.200	.066	-.868	.076	-.462
-.572	-.177	.101	-.635	.297	-.970	.298	-.668	.136	-.785	.136	-.403
-.462	-.206	.185	-.604	.400	-.184	.397	-.261	.214	-.757	.221	-.356
-.329	-.194	.398	-.565	.604	-.198	.501	-.181	.292	-.741	.295	-.318
-.172	-.121	.737	.005	.785	.111	.603	-.070	.403	-.693	.396	-.296
-.030	-.234			.967	.178	.703	.044	.489	-.599	.497	-.249
.128	-.350			1.000	.028	.784	.044	.594	-.544	.597	-.227
.418	-.343					.868	.093	.700	-.428	.702	-.229
.564	-.310					.923	.102	.786	-.328	.786	-.208
.710	-.083					.972	.078	.858	-.152	.864	-.214
.976	.198							.919	.107	.912	-.201
1.072	.222							.967	.136		
1.110	.179										

	.133	.307	.480	.653	.804	.933
CN=	-.1937	-.3680	-.3399	-.3294	-.4824	-.1795
CM=	-.0538	-.0365	-.0772	-.0878	-.0054	-.0314

[redacted]

TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90. Continued.

$\alpha = -2.96^{\circ}$; $C_L = -0.305$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	.034	-.021	.246	.023	.248	.025	.229	.022	.351	.018	.417
	-.567	-.028	.035	.048	.068	.092	.079	.122	.075	.134	.077	.055
	-.452	-.130	.105	-.009	.134	.018	.133	.055	.129	.092	.129	.087
	-.311	-.196	.178	-.055	.209	-.019	.214	.008	.201	.044	.209	.035
	-.023	-.156	.286	-.075	.294	-.047	.295	-.019	.294	.019	.293	-.024
	.133	-.079	.396	-.084	.404	-.068	.407	-.047	.397	-.009	.494	-.103
	.272	-.046	.514	-.077	.497	-.077	.502	-.082	.495	-.073	.590	-.153
	.416	-.010	.618	-.100	.599	-.120	.601	-.123	.594	-.114	.693	-.208
	.565	.014	.733	-.105	.700	-.158	.698	-.184	.693	-.181	.777	-.272
	.713	.010	.835	-.125	.864	-.189	.863	-.250	.784	-.234	.861	-.313
	.854	-.019	.919	-.105	.926	-.135	.923	-.188	.856	-.245	.918	-.377
	.980	-.046	.987	-.009	.975	-.044	.977	-.031	.926	-.167	.972	-.336
	1.074	-.065							.977	.013		
	1.122	-.019										
						LOWER SURFACE						
	-.660	-.019	-.022	-.364	.024	-.954	.025	-1.053	.019	-.911	.020	-.537
	-.616	-.122	.038	-.596	.075	-.967	.130	-1.123	.066	-.879	.076	-.536
	-.572	-.166	.101	-.554	.297	-.540	.298	-.406	.136	-.803	.136	-.453
	-.462	-.184	.185	-.554	.400	-.262	.397	-.257	.214	-.735	.221	-.432
	-.329	-.168	.398	-.395	.604	-.213	.501	-.187	.292	-.690	.295	-.374
	-.172	-.118	.737	.029	.785	.110	.603	-.023	.403	-.602	.396	-.319
	-.030	-.224			.967	.200	.703	.078	.489	-.531	.497	-.256
	.128	-.337			1.000	.036	.784	.105	.594	-.407	.597	-.233
	.418	-.288					.868	.144	.700	-.230	.702	-.191
	.564	-.240					.923	.170	.786	-.086	.786	-.193
	.710	-.048					.972	.127	.858	.026	.864	-.116
	.976	.204							.919	.115	.912	-.157
	1.072	.213							.967	.119		
	1.110	.174										
CN=	-.1186		-.2444		-.2362		-.2270		-.3922		-.1911	
CM=	-.0524		-.0447		-.0752		-.0970		-.0315		-.0390	

$\alpha = -2.00^{\circ}$; $C_L = -0.203$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	.037	-.021	.165	.023	.127	.025	.168	.022	.294	.018	.370
	-.567	-.041	.035	-.025	.068	.008	.079	.045	.075	.072	.077	-.004
	-.452	-.154	.105	-.076	.134	-.047	.133	-.012	.129	.029	.129	.039
	-.311	-.228	.178	-.109	.209	-.081	.214	-.035	.201	.002	.209	-.007
	-.023	-.181	.286	-.135	.294	-.098	.295	-.058	.294	-.029	.293	-.054
	.133	-.115	.396	-.132	.404	-.110	.407	-.077	.397	-.054	.494	-.127
	.272	-.070	.514	-.110	.497	-.122	.502	-.121	.495	-.099	.590	-.153
	.416	-.034	.618	-.124	.599	-.146	.601	-.151	.594	-.142	.693	-.184
	.565	-.012	.733	-.133	.700	-.183	.698	-.202	.693	-.203	.777	-.207
	.713	-.011	.835	-.137	.864	-.209	.863	-.261	.784	-.241	.861	-.185
	.854	-.043	.919	-.115	.926	-.139	.923	-.171	.856	-.253	.918	-.155
	.980	-.066	.987	-.015	.975	-.042	.977	-.023	.926	-.169	.972	-.021
	1.074	-.075							.977	-.007		
	1.122	-.020										
						LOWER SURFACE						
	-.660	.001	-.022	-.221	.024	-.752	.025	-.940	.019	-.987	.020	-.862
	-.616	-.085	.038	-.447	.075	-.771	.130	-1.008	.066	-.919	.076	-.851
	-.572	-.125	.101	-.485	.297	-.321	.298	-.302	.136	-.760	.136	-.650
	-.462	-.166	.185	-.485	.400	-.298	.397	-.276	.214	-.628	.221	-.566
	-.329	-.157	.398	-.305	.604	-.212	.501	-.226	.292	-.497	.295	-.484
	-.172	-.109	.737	.051	.785	.110	.603	-.053	.403	-.381	.396	-.353
	-.030	-.212			.967	.199	.703	.054	.489	-.287	.497	-.228
	.128	-.313			1.000	.033	.784	.106	.594	-.181	.597	-.133
	.418	-.247					.868	.159	.700	-.061	.702	-.046
	.564	-.188					.923	.183	.786	.038	.786	-.029
	.710	-.013					.972	.140	.858	.112	.864	.068
	.976	.216							.919	.139	.912	.099
	1.072	.216							.967	.110		
	1.110	.168										
CN=	-.0328		-.1329		-.1275		-.1662		-.2341		-.2090	
CM=	-.0505		-.0506		-.0724		-.0935		-.0694		-.0597	

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90. Continued.

$\alpha = -1.04^{\circ}$; $C_L = -0.094$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	.021	-.021	.068	.023	-.016	.025	.048	.022	.202	.018	.290
-.567	-.067	.035	-.123	.068	-.087	.079	-.056	.075	-.002	.077	-.090
-.452	-.193	.105	-.147	.134	-.149	.133	-.086	.129	-.044	.129	-.009
-.311	-.267	.178	-.188	.209	-.149	.214	-.095	.201	-.067	.209	-.057
-.023	-.220	.286	-.188	.294	-.141	.295	-.113	.294	-.079	.293	-.101
.133	-.135	.396	-.187	.404	-.161	.407	-.132	.397	-.092	.494	-.154
.272	-.102	.514	-.155	.497	-.166	.502	-.156	.495	-.143	.590	-.171
.416	-.067	.618	-.158	.599	-.179	.601	-.190	.594	-.179	.693	-.190
.565	-.026	.733	-.156	.700	-.213	.698	-.238	.693	-.230	.777	-.206
.713	-.027	.835	-.157	.864	-.220	.863	-.270	.784	-.261	.861	-.175
.854	-.064	.919	-.119	.926	-.147	.923	-.173	.856	-.276	.918	-.137
.980	-.083	.987	-.022	.975	-.052	.977	-.021	.926	-.180	.972	-.004
1.074	-.062							.977	-.016		
1.122	-.027										
					LOWER SURFACE						
-.660	.009	-.022	-.083	.024	-.517	.025	-.765	.019	-.860	.020	-.934
-.616	-.047	.038	-.310	.075	-.595	.130	-.816	.066	-.945	.076	-.890
-.572	-.102	.101	-.380	.297	-.321	.298	-.328	.136	-.883	.136	-.850
-.462	-.134	.185	-.377	.400	-.251	.397	-.300	.214	-.379	.221	-.555
-.329	-.136	.398	-.256	.604	-.154	.501	-.224	.292	-.315	.295	-.352
-.172	-.092	.737	.065	.785	.116	.603	-.056	.403	-.245	.396	-.210
-.030	-.194			.967	.195	.703	.058	.489	-.244	.497	-.115
.128	-.268			1.000	.020	.784	.107	.594	-.178	.597	-.031
.418	-.221					.868	.180	.700	.003	.702	.067
.564	-.150					.923	.186	.786	.112	.786	.130
.710	.014					.972	.144	.858	.178	.864	.152
.976	.223							.919	.201	.912	.160
1.072	.223							.967	.142		
1.110	.173										
CN=	.0551		-.0204		-.0259		-.0850		-.1208		-.1186
CM=	-.0397		-.0521		-.0740		-.0911		-.0891		-.0893

$\alpha = 0.02^{\circ}$; $C_L = 0.024$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	.015	-.021	-.032	.023	-.169	.025	-.079	.022	.081	.018	.190
-.567	-.081	.035	-.227	.068	-.238	.079	-.150	.075	-.122	.077	-.204
-.452	-.219	.105	-.240	.134	-.232	.133	-.171	.129	-.115	.129	-.089
-.311	-.291	.178	-.248	.209	-.213	.214	-.167	.201	-.138	.209	-.127
-.023	-.244	.286	-.253	.294	-.216	.295	-.163	.294	-.140	.293	-.149
.133	-.158	.396	-.233	.404	-.194	.407	-.171	.397	-.146	.494	-.169
.272	-.123	.514	-.186	.497	-.199	.502	-.200	.495	-.185	.590	-.182
.416	-.097	.618	-.188	.599	-.213	.601	-.217	.594	-.209	.693	-.207
.565	-.056	.733	-.170	.700	-.239	.698	-.272	.693	-.258	.777	-.213
.713	-.055	.835	-.167	.864	-.236	.863	-.276	.784	-.285	.861	-.172
.854	-.086	.919	-.119	.926	-.146	.923	-.168	.856	-.282	.918	-.127
.980	-.096	.987	-.015	.975	-.050	.977	-.024	.926	-.173	.972	.012
1.074	-.090							.977	-.010		
1.122	-.030										
					LOWER SURFACE						
-.660	.039	-.022	.032	.024	-.264	.025	-.520	.019	-.659	.020	-.633
-.616	-.039	.038	-.223	.075	-.425	.130	-.343	.066	-.698	.076	-.699
-.572	-.074	.101	-.274	.297	-.285	.298	-.297	.136	-.515	.136	-.665
-.462	-.112	.185	-.288	.400	-.229	.397	-.256	.214	-.315	.221	-.398
-.329	-.117	.398	-.203	.604	-.130	.501	-.195	.292	-.285	.295	-.255
-.172	-.078	.737	.086	.785	.143	.603	-.047	.403	-.223	.396	-.165
-.030	-.173			.967	.199	.703	.065	.489	-.223	.497	-.128
.128	-.257			1.000	.007	.784	.116	.594	-.168	.597	-.044
.418	-.178					.868	.174	.700	.024	.702	.089
.564	-.110					.923	.185	.786	.132	.786	.160
.710	.037					.972	.139	.858	.192	.864	.199
.976	.234							.919	.214	.912	.197
1.072	.223							.967	.162		
1.110	.170										
CN=	.1383		.0651		.0772		.0561		-.0057		-.0125
CM=	-.0364		-.0541		-.0741		-.0831		-.0879		-.0872

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(d) M = 0.90. Continued.

$\alpha = 1.05^o$; $C_L = 0.135$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.003	-.021	-.160	.023	-.337	.025	-.292	.022	-.075	.018	.042
-.567	-.112	.035	-.333	.068	-.516	.079	-.307	.075	-.277	.077	-.327
-.452	-.246	.105	-.358	.134	-.360	.133	-.300	.129	-.231	.129	-.178
-.311	-.333	.178	-.337	.209	-.315	.214	-.250	.201	-.227	.209	-.190
-.023	-.271	.286	-.318	.294	-.270	.295	-.236	.294	-.207	.293	-.206
.133	-.185	.396	-.279	.404	-.252	.407	-.225	.397	-.200	.494	-.195
.272	-.136	.514	-.233	.497	-.243	.502	-.245	.495	-.222	.590	-.207
.416	-.123	.618	-.223	.599	-.248	.601	-.251	.594	-.245	.693	-.227
.565	-.094	.733	-.202	.700	-.263	.698	-.304	.693	-.295	.777	-.235
.713	-.074	.835	-.180	.864	-.242	.863	-.274	.784	-.317	.861	-.196
.854	-.104	.919	-.120	.926	-.149	.923	-.163	.856	-.294	.918	-.150
.980	-.118	.987	-.019	.975	-.060	.977	-.029	.926	-.174	.972	-.011
1.074	-.099							.977	-.024		
1.122	-.041										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.063	-.022	.126	.024	-.075	.025	-.196	.019	-.313	.020	-.323
-.616	-.002	.038	-.109	.075	-.246	.130	-.295	.066	-.456	.076	-.469
-.572	-.046	.101	-.194	.297	-.215	.298	-.217	.136	-.332	.136	-.484
-.462	-.086	.185	-.214	.400	-.197	.397	-.202	.214	-.285	.221	-.394
-.329	-.080	.398	-.162	.604	-.118	.501	-.168	.292	-.212	.295	-.192
-.172	-.055	.737	.103	.785	.160	.603	-.037	.403	-.182	.396	-.167
-.030	-.151			.967	.198	.703	.060	.489	-.196	.497	-.125
.128	-.221			1.000	-.004	.784	.141	.594	-.157	.597	-.041
.418	-.127					.868	.197	.700	.041	.702	.096
.564	-.067					.923	.220	.786	.159	.786	.178
.710	.067					.972	.164	.858	.226	.864	.224
.976	.249							.919	.248	.912	.224
1.072	.228							.967	.157		
1.110	.172										
CN=	.2396		.1915		.1956		.1728		.1186		.0891
CM=	-.0274		-.0564		-.0715		-.0833		-.0892		-.0874

$\alpha = 2.01^o$; $C_L = 0.239$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.008	-.021	-.256	.023	-.561	.025	-.452	.022	-.332	.018	-.138
-.567	-.135	.035	-.458	.068	-.763	.079	-.545	.075	-.452	.077	-.482
-.452	-.271	.105	-.410	.134	-.475	.133	-.393	.129	-.339	.129	-.267
-.311	-.368	.178	-.434	.209	-.419	.214	-.323	.201	-.309	.209	-.262
-.023	-.293	.286	-.390	.294	-.308	.295	-.297	.294	-.248	.293	-.255
.133	-.213	.396	-.355	.404	-.289	.407	-.265	.397	-.246	.494	-.217
.272	-.178	.514	-.266	.497	-.312	.502	-.276	.495	-.269	.590	-.219
.416	-.149	.618	-.235	.599	-.266	.601	-.294	.594	-.286	.693	-.237
.565	-.122	.733	-.210	.700	-.291	.698	-.323	.693	-.321	.777	-.248
.713	-.110	.835	-.186	.864	-.244	.863	-.283	.784	-.328	.861	-.202
.854	-.135	.919	-.120	.926	-.143	.923	-.165	.856	-.294	.918	-.152
.980	-.133	.987	-.013	.975	-.060	.977	-.041	.926	-.167	.972	-.011
1.074	-.109							.977	-.033		
1.122	-.045										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.076	-.022	.204	.024	.076	.025	-.029	.019	-.079	.020	-.133
-.616	.012	.038	-.030	.075	-.145	.130	-.173	.066	-.271	.076	-.300
-.572	-.017	.101	-.100	.297	-.169	.298	-.148	.136	-.245	.136	-.331
-.462	-.059	.185	-.136	.400	-.138	.397	-.162	.214	-.217	.221	-.261
-.329	-.067	.398	-.135	.604	-.105	.501	-.142	.292	-.181	.295	-.231
-.172	-.038	.737	.108	.785	.165	.603	-.022	.403	-.126	.396	-.144
-.030	-.130			.967	.192	.703	.055	.489	-.155	.497	-.117
.128	-.190			1.000	-.010	.784	.143	.594	-.137	.597	-.041
.418	-.095					.868	.214	.700	.048	.702	.098
.564	-.048					.923	.256	.786	.174	.786	.189
.710	.082					.972	.181	.858	.246	.864	.239
.976	.258							.919	.269	.912	.245
1.072	.230							.967	.156		
1.110	.175										
CN=	.3242		.2757		.2958		.2783		.2261		.1770
CM=	-.0216		-.0539		-.0686		-.0835		-.0884		-.0836

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90. Continued.

$\alpha = 2.45^{\circ}$; $C_L = 0.285$

STA .133

X/C	CP
UPPER SURFACE	
-.660	-.017
-.567	-.134
-.452	-.284
-.311	-.360
-.023	-.320
.133	-.223
.272	-.195
.416	-.163
.565	-.128
.713	-.120
.854	-.141
.980	-.138
1.074	-.112
1.122	-.047
LOWER SURFACE	
-.660	.084
-.616	.031
-.572	.000
-.462	-.054
-.329	-.065
-.172	-.033
-.030	-.120
.128	-.184
.418	-.095
.564	-.033
.710	.088
.976	.266
1.072	.238
1.110	.176

CN= .3526
CM= -.0211

STA .307

X/C	CP
UPPER SURFACE	
-.021	-.339
.035	-.544
.105	-.452
.178	-.456
.286	-.410
.396	-.375
.514	-.294
.618	-.250
.733	-.210
.835	-.179
.919	-.109
.987	-.015
LOWER SURFACE	
-.022	.237
.038	.004
.101	-.079
.185	-.103
.398	-.113
.737	.114

CN= .3225
CM= -.0527

STA .480

X/C	CP
UPPER SURFACE	
.023	-.625
.068	-.886
.134	-.573
.209	-.487
.294	-.360
.404	-.280
.497	-.318
.599	-.261
.700	-.295
.864	-.235
.926	-.142
.975	-.059
LOWER SURFACE	
.024	.094
.075	-.105
.297	-.141
.400	-.128
.604	-.093
.785	.168
.967	.195
1.000	-.010

CN= .3373
CM= -.0660

STA .653

X/C	CP
UPPER SURFACE	
.025	-.579
.079	-.673
.133	-.445
.214	-.377
.295	-.313
.407	-.284
.502	-.294
.601	-.310
.698	-.341
.863	-.280
.923	-.164
.977	-.041
LOWER SURFACE	
.025	.046
.130	-.130
.298	-.140
.397	-.138
.501	-.127
.603	-.019
.703	.056
.784	.142
.868	.222
.923	.266
.972	.185

CN= .3267
CM= -.0822

STA .804

X/C	CP
UPPER SURFACE	
.022	-.416
.075	-.536
.129	-.400
.201	-.350
.294	-.284
.397	-.264
.495	-.285
.594	-.293
.693	-.324
.784	-.334
.856	-.293
.926	-.159
.977	-.030
LOWER SURFACE	
.019	-.010
.066	-.191
.136	-.176
.214	-.180
.292	-.157
.403	-.111
.489	-.142
.594	-.124
.700	.056
.786	.177
.858	.245
.919	.269
.967	.153

CN= .2723
CM= -.0862

STA .933

X/C	CP
UPPER SURFACE	
.018	-.260
.077	-.567
.129	-.349
.209	-.289
.293	-.291
.494	-.222
.590	-.222
.693	-.239
.777	-.247
.861	-.200
.918	-.146
.972	-.011
LOWER SURFACE	
.020	-.001
.076	-.276
.136	-.281
.221	-.252
.295	-.213
.396	-.126
.497	-.111
.597	-.035
.702	.098
.786	.187
.864	.237
.912	.242

CN= .2171
CM= -.0791

$\alpha = 2.94^{\circ}$; $C_L = 0.341$

STA .133

X/C	CP
UPPER SURFACE	
-.660	-.029
-.567	-.155
-.452	-.300
-.311	-.389
-.023	-.345
.133	-.253
.272	-.205
.416	-.187
.565	-.147
.713	-.138
.854	-.161
.980	-.153
1.074	-.119
1.122	-.051
LOWER SURFACE	
-.660	.086
-.616	.044
-.572	.013
-.462	-.036
-.329	-.052
-.172	-.029
-.030	-.110
.128	-.172
.418	-.077
.564	-.018
.710	.103
.976	.268
1.072	.241
1.110	.180

CN= .4096
CM= -.0160

STA .307

X/C	CP
UPPER SURFACE	
-.021	-.447
.035	-.717
.105	-.503
.178	-.505
.286	-.448
.396	-.424
.514	-.339
.618	-.294
.733	-.221
.835	-.176
.919	-.104
.987	-.018
LOWER SURFACE	
-.022	.294
.038	.044
.101	-.035
.185	-.080
.398	-.095
.737	.122

CN= .3684
CM= -.0529

STA .480

X/C	CP
UPPER SURFACE	
.023	-.815
.068	-.901
.134	-.869
.209	-.612
.294	-.437
.404	-.273
.497	-.312
.599	-.263
.700	-.301
.864	-.243
.926	-.142
.975	-.062
LOWER SURFACE	
.024	.189
.075	-.046
.297	-.123
.400	-.099
.604	-.085
.785	.177
.967	.193
1.000	-.016

CN= .4086
CM= -.0619

STA .653

X/C	CP
UPPER SURFACE	
.025	-.725
.079	-.866
.133	-.628
.214	-.341
.295	-.330
.407	-.323
.502	-.318
.601	-.329
.698	-.353
.863	-.279
.923	-.161
.977	-.052
LOWER SURFACE	
.025	.140
.130	-.092
.298	-.119
.397	-.120
.501	-.103
.603	-.010
.703	.063
.784	.149
.868	.232
.923	.282
.972	.184

CN= .3893
CM= -.0809

STA .804

X/C	CP
UPPER SURFACE	
.022	-.513
.075	-.715
.129	-.450
.201	-.420
.294	-.335
.397	-.292
.495	-.317
.594	-.322
.693	-.344
.784	-.340
.856	-.284
.926	-.155
.977	-.032
LOWER SURFACE	
.019	.064
.066	-.120
.136	-.138
.214	-.133
.292	-.117
.403	-.094
.489	-.131
.594	-.113
.700	.059
.786	.179
.858	.248
.919	.273
.967	.153

CN= .3333
CM= -.0855

STA .933

X/C	CP
UPPER SURFACE	
.018	-.399
.077	-.764
.129	-.425
.209	-.349
.293	-.334
.494	-.233
.590	-.232
.693	-.241
.777	-.252
.861	-.203
.918	-.149
.972	-.012
LOWER SURFACE	
.020	.077
.076	-.199
.136	-.224
.221	-.188
.295	-.181
.396	-.123
.497	-.106
.597	-.035
.702	.098
.786	.187
.864	.235
.912	.240

CN= .2749
CM= -.0742

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90. Continued.

$\alpha = 3.47^{\circ}$; $C_L = 0.403$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.039	-.021	-.544	.023	-.870	.025	-.874	.022	-.729	.018	-.623
-.567	-.177	.035	-.791	.068	-.996	.079	-.963	.075	-.789	.077	-.839
-.452	-.313	.105	-.548	.134	-.949	.133	-.932	.129	-.567	.129	-.499
-.311	-.400	.178	-.568	.209	-.744	.214	-.479	.201	-.396	.209	-.382
-.023	-.335	.286	-.476	.294	-.517	.295	-.311	.294	-.367	.293	-.345
.133	-.264	.396	-.464	.404	-.297	.407	-.300	.397	-.333	.494	-.224
.272	-.211	.514	-.376	.497	-.296	.502	-.310	.495	-.340	.590	-.237
.416	-.198	.618	-.341	.599	-.257	.601	-.324	.594	-.344	.693	-.250
.565	-.166	.733	-.218	.700	-.283	.698	-.360	.693	-.345	.777	-.259
.713	-.148	.835	-.168	.864	-.235	.863	-.283	.784	-.342	.861	-.207
.854	-.185	.919	-.087	.926	-.136	.923	-.172	.856	-.282	.918	-.154
.980	-.165	.987	-.017	.975	-.060	.977	-.053	.926	-.151	.972	-.017
1.074	-.126							.977	-.033		
1.122	-.059										
				LOWER SURFACE							
-.660	.093	-.022	.316	.024	.229	.025	.188	.019	.187	.020	.166
-.616	.057	.038	.082	.075	.004	.130	-.048	.066	-.054	.076	-.090
-.572	.028	.101	.008	.297	-.093	.298	-.091	.136	-.091	.136	-.170
-.462	-.021	.185	-.048	.400	-.091	.397	-.100	.214	-.088	.221	-.166
-.329	-.040	.398	-.068	.604	-.077	.501	-.090	.292	-.096	.295	-.143
-.172	-.016	.737	.128	.785	.180	.603	-.000	.403	-.073	.396	-.109
-.030	-.091			.967	.196	.703	.066	.489	-.109	.497	-.098
.128	-.158			1.000	-.014	.784	.154	.594	-.100	.597	-.034
.418	-.057					.868	.242	.700	.064	.702	.095
.564	-.005					.923	.287	.786	.179	.786	.176
.710	.110					.972	.193	.858	.248	.864	.232
.976	.279							.919	.269	.912	.232
1.072	.241							.967	.151		
1.110	.181										
CN=	.4507		.4440		.4544		.4503		.3883		.3199
CM=	-.0146		-.0528		-.0575		-.0768		-.0825		-.0682

$\alpha = 3.97^{\circ}$; $C_L = 0.461$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.051	-.021	-.806	.023	-.951	.025	-.944	.022	-.823	.018	-.733
-.567	-.188	.035	-.799	.066	-1.076	.079	-1.031	.075	-.952	.077	-.916
-.452	-.334	.105	-.620	.134	-1.002	.133	-1.013	.129	-.893	.129	-.795
-.311	-.411	.178	-.608	.209	-.983	.214	-.963	.201	-.465	.209	-.397
-.023	-.344	.286	-.566	.294	-.675	.295	-.545	.294	-.352	.293	-.344
.133	-.273	.396	-.494	.404	-.409	.407	-.283	.397	-.300	.494	-.228
.272	-.224	.514	-.402	.497	-.284	.502	-.267	.495	-.332	.590	-.237
.416	-.213	.618	-.362	.599	-.245	.601	-.296	.594	-.345	.693	-.255
.565	-.188	.733	-.231	.700	-.279	.698	-.340	.693	-.347	.777	-.261
.713	-.183	.835	-.153	.864	-.220	.863	-.291	.784	-.332	.861	-.209
.854	-.219	.919	-.083	.926	-.132	.923	-.180	.856	-.273	.918	-.150
.980	-.181	.987	-.019	.975	-.053	.977	-.057	.926	-.149	.972	-.017
1.074	-.135							.977	-.034		
1.122	-.064										
				LOWER SURFACE							
-.660	.100	-.022	.347	.024	.277	.025	.218	.019	.213	.020	.224
-.616	.072	.038	.143	.075	.046	.130	-.031	.066	.001	.076	-.047
-.572	.043	.101	.041	.297	-.065	.298	-.065	.136	-.052	.136	-.115
-.462	-.007	.185	-.012	.400	-.079	.397	-.086	.214	-.063	.221	-.128
-.329	-.020	.398	-.058	.604	-.070	.501	-.077	.292	-.064	.295	-.132
-.172	-.006	.737	.133	.785	.181	.603	.016	.403	-.047	.396	-.107
-.030	-.080			.967	.200	.703	.073	.489	-.092	.497	-.099
.128	-.128			1.000	-.016	.784	.157	.594	-.090	.597	-.034
.418	-.037					.868	.244	.700	.068	.702	.092
.564	.007					.923	.295	.786	.180	.786	.177
.710	.127					.972	.196	.858	.254	.864	.226
.976	.286							.919	.271	.912	.231
1.072	.249							.967	.148		
1.110	.184										
CN=	.5090		.5038		.5232		.5291		.4396		.3635
CM=	-.0154		-.0501		-.0546		-.0732		-.0762		-.0621

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(d) M = 0.90. Continued.

$\alpha = 4.94^{\circ}$; $C_L = 0.580$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.065	-.021	-1.008	.023	-1.064	.025	-1.049	.022	-.989	.018	-.917
-.567	-.202	.035	-1.106	.068	-1.201	.079	-1.147	.075	-1.075	.077	-1.092
-.452	-.342	.105	-.673	.134	-1.057	.133	-1.101	.129	-1.059	.129	-1.032
-.311	-.436	.178	-.634	.209	-1.039	.214	-1.063	.201	-1.049	.209	-.990
-.023	-.354	.286	-.616	.294	-.982	.295	-1.022	.294	-.799	.293	-.433
.133	-.294	.396	-.590	.404	-.620	.407	-.543	.397	-.430	.494	-.157
.272	-.247	.514	-.478	.497	-.432	.502	-.465	.495	-.362	.590	-.170
.416	-.240	.618	-.453	.599	-.276	.601	-.288	.594	-.293	.693	-.223
.569	-.210	.733	-.272	.700	-.229	.698	-.252	.693	-.274	.777	-.237
.713	-.215	.835	-.146	.864	-.184	.863	-.264	.784	-.268	.861	-.210
.854	-.271	.919	-.073	.926	-.105	.923	-.167	.856	-.214	.918	-.163
.980	-.229	.987	-.029	.975	-.047	.977	-.047	.926	-.129	.972	-.036
1.074	-.157							.977	-.037		
1.122	-.073										
				LOWER SURFACE							
-.660	.114	-.022	.390	.024	.346	.025	.298	.019	.303	.020	.315
-.616	.099	.038	.196	.075	.114	.130	.035	.066	.068	.076	.028
-.572	.074	.101	.098	.297	-.019	.298	-.024	.136	.009	.136	-.047
-.462	.025	.185	.033	.400	-.036	.397	-.051	.214	-.020	.221	-.081
-.329	.013	.398	-.018	.604	-.058	.501	-.051	.292	-.029	.295	-.098
-.172	.022	.737	.143	.785	.190	.603	.034	.403	-.026	.396	-.090
-.030	-.053			.967	.200	.703	.085	.489	-.068	.497	-.083
.128	-.112			1.000	-.017	.784	.168	.594	-.067	.597	-.024
.418	-.011					.868	.257	.700	.080	.702	.088
.564	.037					.923	.317	.786	.189	.786	.170
.710	.150					.972	.209	.858	.263	.864	.220
.976	.298							.919	.273	.912	.226
1.072	.259							.967	.156		
1.110	.185										
CN=	.5941		.6114		.6336		.6639		.5788		.4619
CM=	-.0167		-.0533		-.0545		-.0752		-.0645		-.0468

$\alpha = 6.06^{\circ}$; $C_L = 0.706$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.086	-.021	-1.150	.023	-1.173	.025	-1.146	.022	-1.099	.018	-.980
-.567	-.235	.035	-1.263	.068	-1.332	.079	-1.232	.075	-1.197	.077	-.951
-.452	-.373	.105	-.860	.134	-1.200	.133	-1.176	.129	-1.163	.129	-.844
-.311	-.460	.178	-.727	.209	-1.172	.214	-1.149	.201	-1.141	.209	-.698
-.023	-.382	.286	-.654	.294	-1.105	.295	-1.098	.294	-.798	.293	-.614
.133	-.301	.396	-.617	.404	-.856	.407	-.812	.397	-.586	.494	-.391
.272	-.266	.514	-.549	.497	-.628	.502	-.615	.495	-.566	.590	-.277
.416	-.257	.618	-.513	.599	-.486	.601	-.511	.594	-.498	.693	-.209
.565	-.230	.733	-.377	.700	-.253	.698	-.318	.693	-.400	.777	-.168
.713	-.250	.835	-.151	.864	-.104	.863	-.171	.784	-.267	.861	-.131
.854	-.327	.919	-.077	.926	-.082	.923	-.126	.856	-.216	.918	-.129
.980	-.329	.987	-.045	.975	-.044	.977	-.034	.926	-.161	.972	-.099
1.074	-.202							.977	-.118		
1.122	-.105										
				LOWER SURFACE							
-.660	.127	-.022	.437	.024	.414	.025	.369	.019	.371	.020	.380
-.616	.116	.038	.258	.075	.179	.130	.086	.066	.148	.076	.099
-.572	.113	.101	.159	.297	.022	.298	.014	.136	.067	.136	.008
-.462	.054	.185	.091	.400	-.012	.397	-.023	.214	.030	.221	-.045
-.329	.042	.398	.039	.604	-.042	.501	-.029	.292	.013	.295	-.076
-.172	.054	.737	.161	.785	.192	.603	.049	.403	.003	.396	-.080
-.030	-.020			.967	.197	.703	.093	.489	-.046	.497	-.097
.128	-.066			1.000	-.031	.784	.166	.594	-.049	.597	-.047
.418	.021					.868	.264	.700	.080	.702	.064
.564	.071					.923	.325	.786	.182	.786	.137
.710	.172					.972	.220	.858	.253	.864	.185
.976	.317							.919	.266	.912	.181
1.072	.262							.967	.122		
1.110	.185										
CN=	.7037		.7303		.7583		.7789		.7018		.4816
CM=	-.0196		-.0620		-.0617		-.0851		-.0819		-.0473

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(d) M = 0.90. Continued.

$\alpha = 7.15^o$; $C_L = 0.801$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.120	-.021	-1.251	.023	-1.240	.025	-1.237	.022	-1.201	.018	-.659
-.567	-.274	.035	-1.382	.068	-1.379	.079	-1.303	.075	-1.278	.077	-.607
-.452	-.406	.105	-1.031	.134	-1.281	.133	-1.270	.129	-1.256	.129	-.563
-.311	-.490	.178	-.964	.209	-1.258	.214	-1.231	.201	-1.234	.209	-.547
-.023	-.419	.266	-.787	.294	-1.202	.295	-1.172	.294	-.836	.293	-.509
.133	-.331	.396	-.662	.404	-.858	.407	-.802	.397	-.681	.494	-.423
.272	-.291	.514	-.570	.497	-.726	.502	-.734	.495	-.636	.590	-.383
.416	-.290	.618	-.548	.599	-.647	.601	-.586	.594	-.584	.693	-.332
.565	-.262	.733	-.462	.700	-.523	.698	-.307	.693	-.496	.777	-.303
.713	-.277	.835	-.166	.864	-.179	.863	-.189	.784	-.426	.861	-.267
.854	-.363	.919	-.085	.926	-.143	.923	-.189	.856	-.345	.918	-.259
.980	-.387	.967	-.048	.975	-.124	.977	-.106	.926	-.289	.972	-.233
1.074	-.250							.977	-.232		
1.122	-.126										
					LOWER SURFACE						
-.660	.134	-.022	.451	.024	.461	.025	.432	.019	.405	.020	.415
-.616	.153	.038	.318	.075	.250	.130	.122	.066	.186	.076	.139
-.572	.141	.101	.213	.297	.059	.258	.034	.136	.093	.136	.042
-.462	.068	.185	.137	.400	.022	.397	-.018	.214	.044	.221	-.018
-.329	.006	.398	.061	.604	-.026	.501	-.042	.292	.018	.295	-.052
-.172	.083	.737	.171	.785	.199	.603	.029	.403	.004	.396	-.083
-.030	.017			.967	.192	.703	.048	.489	-.036	.497	-.109
.128	-.033			1.000	-.038	.784	.126	.594	-.049	.597	-.076
.418	.055					.868	.234	.700	.084	.702	.022
.564	.093					.923	.294	.786	.176	.786	.092
.710	.191					.972	.170	.858	.241	.864	.141
.976	.329							.919	.245	.912	.138
1.072	.268							.967	.100		
1.110	.188										
CN=	.6116		.8432		.8906		.6289		.8026		.4549
CM=	-.0142		-.0666		-.0948		-.0831		-.1079		-.0771

$\alpha = 8.17^o$; $C_L = 0.863$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.150	-.021	-1.321	.023	-1.324	.025	-1.294	.022	-1.251	.018	-.665
-.567	-.323	.035	-1.440	.068	-1.436	.079	-1.345	.075	-1.327	.077	-.623
-.452	-.436	.105	-1.118	.134	-1.365	.133	-1.323	.129	-1.305	.129	-.578
-.311	-.520	.178	-1.076	.209	-1.324	.214	-1.184	.201	-1.253	.209	-.556
-.023	-.458	.266	-1.048	.294	-1.017	.295	-.910	.294	-.821	.293	-.525
.133	-.349	.396	-.920	.404	-.812	.407	-.746	.397	-.727	.494	-.450
.272	-.319	.514	-.732	.497	-.800	.502	-.515	.495	-.672	.590	-.403
.416	-.312	.618	-.575	.599	-.751	.601	-.340	.594	-.613	.693	-.363
.565	-.285	.733	-.489	.700	-.682	.698	-.283	.693	-.554	.777	-.337
.713	-.293	.835	-.199	.864	-.461	.863	-.255	.784	-.455	.861	-.308
.854	-.369	.919	-.082	.926	-.415	.923	-.261	.856	-.396	.918	-.291
.980	-.431	.967	-.037	.975	-.319	.977	-.262	.926	-.312	.972	-.279
1.074	-.268							.977	-.259		
1.122	-.140										
					LOWER SURFACE						
-.660	.141	-.022	.462	.024	.507	.025	.450	.019	.437	.020	.443
-.616	.170	.038	.354	.075	.293	.130	.158	.066	.209	.076	.166
-.572	.169	.101	.265	.297	.056	.298	.041	.136	.112	.136	.064
-.462	.121	.185	.191	.400	.045	.397	-.013	.214	.063	.221	-.004
-.329	.101	.398	.092	.604	-.029	.501	-.047	.292	.029	.295	-.041
-.172	.109	.737	.180	.785	.163	.603	.006	.403	-.003	.396	-.076
-.030	.033			.967	.076	.703	.015	.489	-.049	.497	-.101
.128	-.002			1.000	-.382	.784	.096	.594	-.066	.597	-.081
.418	.081					.868	.212	.700	.077	.702	.012
.564	.118					.923	.266	.786	.179	.786	.091
.710	.215					.972	.096	.858	.251	.864	.140
.976	.337							.919	.254	.912	.138
1.072	.273							.967	.095		
1.110	.192										
CN=	.9065		.9782		.9787		.7661		.8394		.4854
CM=	-.0047		-.0774		-.1237		-.0683		-.1153		-.0848

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95.

$\alpha = -4.93^{\circ}$; $C_L = -0.524$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.030	-.021	.362	.023	.362	.025	.339	.022	.433	.018	.467
-.567	.005	.035	.187	.068	.203	.079	.206	.075	.240	.077	.227
-.452	-.095	.105	.111	.134	.111	.133	.140	.129	.177	.129	.158
-.311	-.178	.178	.047	.209	.060	.214	.095	.201	.112	.209	.076
-.023	-.093	.286	.008	.294	.037	.295	.072	.294	.073	.293	.012
.133	-.015	.396	.001	.404	.008	.407	.012	.397	.034	.494	-.131
.272	.015	.514	-.005	.497	-.026	.502	-.030	.495	-.031	.590	-.215
.416	.049	.618	-.043	.599	-.065	.601	-.083	.594	-.104	.693	-.311
.565	.078	.733	-.069	.700	-.113	.698	-.171	.693	-.188	.777	-.422
.713	.071	.835	-.106	.864	-.207	.863	-.270	.784	-.287	.861	-.530
.854	.020	.919	-.093	.926	-.178	.923	-.269	.856	-.411	.918	-.659
.980	-.028	.987	.010	.975	-.052	.977	-.165	.926	-.575	.972	-.270
1.074	-.070							.977	-.537		
1.122	-.054										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.054	-.022	-.626	.024	-.995	.025	-1.080	.019	-.874	.020	-.350
-.616	-.159	.038	-.882	.075	-1.017	0.000	0.000	.066	-.765	.076	-.399
0.000	0.000	.101	-.814	.297	-1.004	.130	-1.125	.136	-.548	.136	-.378
-.329	-.240	.185	-.612	.400	-.973	.298	-.836	.214	-.551	.221	-.321
-.172	-.104	.398	-.620	.604	-.105	.397	-.621	.292	-.495	.295	-.342
-.030	-.225	.737	-.160	.785	.052	.501	-.340	.403	-.533	.396	-.285
.128	-.357			.967	.011	.603	-.312	.489	-.464	.497	-.274
.418	-.302			1.000	-.004	.703	-.060	.594	-.465	.597	-.283
.564	-.353					.784	-.012	.700	-.463	.702	-.223
.710	-.250					.868	-.041	.786	-.417	.786	-.234
.976	.142					.923	-.175	.858	-.380	.864	-.218
1.072	.135							.919	-.376	.912	-.227
1.110	.162							.967	-.383		
0.000	0.000										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	-.2514	-.5225	-.4936	-.4636	-.4236	-.1463
CM=	.0000	-.0010	-.0422	-.0470	.0041	-.0548

$\alpha = -3.97^{\circ}$; $C_L = -0.440$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.046	-.021	.325	.023	.295	.025	.289	.022	.394	.018	.450
-.567	.002	.035	.125	.068	.132	.079	.151	.075	.203	.077	.199
-.452	-.103	.105	.048	.134	.059	.133	.100	.129	.139	.129	.129
-.311	-.199	.178	.000	.209	.023	.214	.044	.201	.082	.209	.056
-.023	-.113	.286	-.019	.294	-.007	.295	.032	.294	.050	.293	.001
.133	-.041	.396	-.040	.404	-.029	.407	-.009	.397	.014	.494	-.126
.272	-.000	.514	-.034	.497	-.060	.502	-.056	.495	-.043	.590	-.204
.416	.032	.618	-.065	.599	-.085	.601	-.097	.594	-.095	.693	-.302
.565	.055	.733	-.083	.700	-.136	.698	-.175	.693	-.171	.777	-.413
.713	.051	.835	-.121	.864	-.198	.863	-.252	.784	-.261	.861	-.515
.854	.007	.919	-.098	.926	-.152	.923	-.204	.856	-.344	.918	-.646
.980	-.033	.987	.008	.975	-.027	.977	-.033	.926	-.515	.972	-.267
1.074	-.072							.977	-.538		
1.122	-.033										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.043	-.022	-.528	.024	-.889	.025	-1.025	.019	-.824	.020	-.405
-.616	-.132	.038	-.746	.075	-.931	0.000	0.000	.066	-.701	.076	-.391
0.000	0.000	.101	-.626	.297	-.937	.130	-1.096	.136	-.720	.136	-.367
-.329	-.234	.185	-.521	.400	-.894	.298	-.879	.214	-.714	.221	-.368
-.172	-.101	.398	-.592	.604	-.188	.397	-.528	.292	-.660	.295	-.345
-.030	-.215	.737	-.097	.785	.040	.501	-.380	.403	-.582	.396	-.311
.128	-.345			.967	.067	.603	-.214	.489	-.522	.497	-.281
.418	-.289			1.000	.034	.703	.044	.594	-.528	.597	-.257
.564	-.310					.784	.055	.700	-.422	.702	-.237
.710	-.189					.868	.058	.786	-.363	.786	-.232
.976	.162					.923	.006	.858	-.305	.864	-.217
1.072	.200							.919	-.284	.912	-.209
1.110	.163							.967	-.288		
0.000	0.000										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	-.1909	-.4113	-.4398	-.4042	-.4507	-.1497
CM=	-.0081	-.0114	-.0382	-.0587	.0019	-.0531

[redacted]

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TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Continued.

$\alpha = -3.03^{\circ}$; $C_L = -0.346$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
					UPPER SURFACE						
-.660	.042	-.021	.255	.023	.208	.025	.217	.022	.339	.018	.420
-.567	-.007	.035	.068	.068	.057	.079	.101	.075	.131	.077	.162
-.452	-.129	.105	-.012	.134	-.011	.133	.045	.129	.088	.129	.095
-.311	-.222	.178	-.049	.209	-.041	.214	.021	.201	.055	.209	.033
-.023	-.125	.286	-.083	.294	-.061	.295	.009	.294	.010	.293	-.023
.133	-.065	.396	-.070	.404	-.068	.407	-.043	.397	-.016	.494	-.131
.272	-.041	.514	-.073	.497	-.084	.502	-.078	.495	-.073	.590	-.200
.416	.003	.618	-.093	.599	-.116	.601	-.124	.594	-.120	.693	-.274
.565	.036	.733	-.114	.700	-.152	.698	-.197	.693	-.201	.777	-.349
.713	.030	.835	-.138	.864	-.205	.863	-.253	.784	-.265	.861	-.382
.854	-.025	.919	-.105	.926	-.134	.923	-.191	.856	-.311	.918	-.578
.980	-.056	.987	.006	.975	-.026	.977	-.034	.926	-.215	.972	-.346
1.074	-.082							.977	-.093		
1.122	-.034										
					LOWER SURFACE						
-.660	-.017	-.022	-.415	.024	-.779	.025	-.921	.019	-.993	.020	-.559
-.616	-.101	.038	-.5a1	.075	-.800	0.000	0.000	.066	-1.083	.076	-.539
C.000	0.000	.101	-.53c	.297	-.816	.130	-1.045	.136	-1.027	.136	-.478
-.329	-.196	.185	-.455	.400	-.724	.298	-.955	.214	-.968	.221	-.483
-.172	-.103	.398	-.547	.604	-.182	.397	-.494	.292	-.759	.295	-.432
-.030	-.211	.737	-.021	.785	.115	.501	-.262	.403	-.614	.396	-.354
.128	-.333			.967	.162	.603	-.038	.489	-.557	.497	-.331
.418	-.276			1.000	.026	.703	.171	.594	-.464	.597	-.287
.564	-.281					.784	.133	.700	-.221	.702	-.275
.710	-.128					.868	.152	.786	-.111	.786	-.265
.976	.176					.923	.174	.858	.060	.864	-.239
1.072	.205							.919	.140	.912	-.219
1.110	.170							.967	.177		
C.000	0.000										
CN=	-.1184		-.2905		-.3100		-.3077		-.4332		-.2169
CM=	-.0137		-.0268		-.0554		-.0853		-.0489		-.0397

$\alpha = -2.05^{\circ}$; $C_L = -0.230$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
					UPPER SURFACE						
-.660	.045	-.021	.184	.023	.113	.025	.147	.022	.277	.018	.360
-.567	-.031	.035	-.014	.068	-.032	.079	.011	.075	.070	.077	.099
-.452	-.150	.105	-.067	.134	-.072	.133	-.025	.129	.020	.129	.035
-.311	-.240	.178	-.124	.209	-.105	.214	-.047	.201	-.016	.209	-.015
-.023	-.178	.286	-.134	.294	-.102	.295	-.062	.294	-.035	.293	-.067
.133	-.092	.396	-.128	.404	-.117	.407	-.083	.397	-.060	.494	-.175
.272	-.053	.514	-.109	.497	-.131	.502	-.113	.495	-.121	.590	-.236
.416	-.025	.618	-.126	.599	-.145	.601	-.150	.594	-.147	.693	-.213
.565	.007	.733	-.137	.700	-.191	.698	-.223	.693	-.215	.777	-.199
.713	.002	.835	-.149	.864	-.224	.863	-.292	.784	-.310	.861	-.178
.854	-.036	.919	-.110	.926	-.138	.923	-.187	.856	-.321	.918	-.207
.980	-.075	.987	.002	.975	-.046	.977	-.030	.926	-.171	.972	-.100
1.074	-.085							.977	-.017		
1.122	-.026										
					LOWER SURFACE						
-.660	.004	-.022	-.20C	.024	-.630	.025	-.817	.019	-.876	.020	-.712
-.616	-.081	.038	-.472	.075	-.671	0.000	0.000	.066	-1.030	.076	-.694
C.000	0.000	.101	-.358	.297	-.719	.130	-.939	.136	-.954	.136	-.556
-.329	-.168	.185	-.455	.400	-.433	.298	-.692	.214	-.945	.221	-.473
-.172	-.101	.398	-.483	.604	-.187	.397	-.385	.292	-.726	.295	-.415
-.030	-.217	.737	.023	.785	.131	.501	-.212	.403	-.520	.396	-.347
.128	-.328			.967	.204	.603	-.032	.489	-.319	.497	-.296
.418	-.251			1.000	.023	.703	.118	.594	-.165	.597	-.234
.564	-.264					.784	.146	.700	-.023	.702	-.199
.710	-.087					.868	.183	.786	.081	.786	-.212
.976	.195					.923	.201	.858	.127	.864	-.142
1.072	.219							.919	.150	.912	-.136
1.110	.172							.967	.110		
C.000	0.000										
CN=	-.0488		-.1813		-.1749		-.1905		-.2838		-.2360
CM=	-.0155		-.0357		-.0678		-.0915		-.0807		-.0217

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Continued.

$\alpha = -1.10^{\circ}$; $C_L = -0.111$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.036	-.021	.113	.023	-.029	.025	.032	.022	.201	.018	.286
-.567	-.038	.035	-.095	.068	-.123	.079	-.089	.075	.000	.077	.013
-.452	-.166	.105	-.124	.134	-.155	.133	-.092	.129	-.039	.129	-.015
-.311	-.263	.178	-.191	.209	-.153	.214	-.107	.201	-.064	.209	-.074
-.023	-.339	.286	-.168	.294	-.169	.295	-.101	.294	-.075	.293	-.109
.133	-.084	.396	-.190	.404	-.142	.407	-.124	.397	-.091	.494	-.196
.272	-.072	.514	-.134	.497	-.187	.502	-.152	.495	-.155	.590	-.268
.416	-.048	.618	-.159	.599	-.171	.601	-.187	.594	-.215	.693	-.338
.565	-.017	.733	-.167	.700	-.222	.698	-.251	.693	-.209	.777	-.113
.713	-.015	.835	-.169	.864	-.245	.863	-.299	.784	-.299	.861	-.124
.854	-.063	.919	-.112	.926	-.149	.923	-.186	.856	-.333	.918	-.108
.980	-.095	.937	-.001	.975	-.045	.977	-.020	.926	-.200	.972	.019
1.074	-.090							.977	-.015		
1.122	-.018										
					LOWER SURFACE						
-.660	.030	-.022	-.052	.024	-.456	.025	-.671	.019	-.736	.020	-.809
-.616	-.048	.038	-.312	.075	-.587	0.000	0.000	.066	-.909	.076	-.851
0.000	0.000	.101	-.365	.297	-.464	.130	-.768	.136	-.854	.136	-.771
-.329	-.159	.135	-.407	.400	-.223	.298	-.325	.214	-.687	.221	-.756
-.172	-.085	.398	-.420	.604	-.215	.397	-.288	.292	-.494	.295	-.550
-.030	-.208	.737	.075	.785	.141	.501	-.262	.403	-.272	.396	-.421
.128	-.321			.967	.215	.603	-.031	.489	-.235	.497	-.258
.418	-.214			1.000	.025	.703	.061	.594	-.182	.597	-.080
.564	-.232					.784	.128	.700	-.012	.702	.035
.710	-.019					.868	.199	.786	.078	.786	.095
.976	.224					.923	.212	.858	.166	.864	.168
1.072	.232							.919	.191	.912	.194
1.110	.182							.967	.140		
0.000	0.000										
CN=	.0541		-.0745		-.0376		-.0666		-.1501		-.1669
CM=	-.0171		-.0480		-.0741		-.0884		-.0896		-.0818

$\alpha = -0.09^{\circ}$; $C_L = 0.012$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.030	-.021	.018	.023	-.141	.025	-.104	.022	.094	.018	.190
-.567	-.065	.035	-.189	.068	-.272	.079	-.214	.075	-.106	.077	-.068
-.452	-.195	.105	-.229	.134	-.261	.133	-.213	.129	-.125	.129	-.093
-.311	-.287	.178	-.273	.209	-.232	.214	-.177	.201	-.131	.209	-.139
-.023	-.378	.286	-.235	.294	-.207	.295	-.158	.294	-.138	.293	-.156
.133	-.119	.396	-.243	.404	-.184	.407	-.171	.397	-.143	.494	-.222
.272	-.095	.514	-.197	.497	-.234	.502	-.199	.495	-.196	.590	-.289
.416	-.081	.618	-.200	.599	-.227	.601	-.221	.594	-.254	.693	-.354
.565	-.039	.733	-.202	.700	-.266	.698	-.296	.693	-.317	.777	-.111
.713	-.040	.835	-.183	.864	-.268	.863	-.338	.784	-.317	.861	-.137
.854	-.100	.919	-.125	.926	-.155	.923	-.159	.856	-.329	.918	-.106
.980	-.123	.987	-.005	.975	-.052	.977	-.016	.926	-.205	.972	.027
1.074	-.104							.977	-.022		
1.122	-.029										
					LOWER SURFACE						
-.660	.042	-.022	.040	.024	-.289	.025	-.491	.019	-.568	.020	-.573
-.616	-.024	.038	-.208	.075	-.451	0.000	0.000	.066	-.743	.076	-.791
0.000	0.000	.101	-.290	.297	-.241	.130	-.549	.136	-.682	.136	-.731
-.329	-.121	.185	-.312	.400	-.259	.298	-.237	.214	-.340	.221	-.621
-.172	-.071	.398	-.225	.604	-.207	.397	-.235	.292	-.278	.295	-.383
-.030	-.192	.737	.088	.785	.142	.501	-.287	.403	-.251	.396	-.282
.128	-.314			.967	.209	.603	-.029	.489	-.184	.497	-.083
.418	-.196			1.000	.016	.703	.067	.594	-.201	.597	-.002
.564	-.148					.784	.113	.700	.014	.702	.102
.710	.033					.868	.177	.786	.116	.786	.164
.976	.237					.923	.185	.858	.174	.864	.204
1.072	.233							.919	.212	.912	.212
1.110	.184							.967	.150		
0.000	0.000										
CN=	.1469		.0777		.0759		.0428		-.0069		-.0343
CM=	-.0213		-.0580		-.0750		-.0838		-.0955		-.0964

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Continued.

$\alpha = 0.99^{\circ}$; $C_L = 0.140$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.010	-.021	-.110	.023	-.405	.025	-.294	.022	-.099	.018	.037
-.567	-.081	.035	-.332	.068	-.494	.079	-.373	.075	-.258	.077	-.204
-.452	-.222	.105	-.303	.134	-.365	.133	-.319	.129	-.235	.129	-.200
-.311	-.311	.178	-.362	.209	-.352	.214	-.281	.201	-.235	.209	-.208
-.023	-.405	.286	-.302	.294	-.339	.295	-.250	.294	-.198	.293	-.217
.133	-.239	.396	-.295	.404	-.272	.407	-.163	.397	-.193	.494	-.291
.272	-.086	.514	-.262	.497	-.282	.502	-.196	.495	-.247	.590	-.326
.416	-.103	.618	-.282	.599	-.266	.601	-.246	.594	-.301	.693	-.229
.565	-.077	.733	-.266	.700	-.227	.698	-.331	.693	-.381	.777	-.145
.713	-.079	.835	-.199	.864	-.275	.863	-.418	.784	-.382	.861	-.169
.854	-.139	.919	-.106	.926	-.167	.923	-.149	.856	-.305	.918	-.130
.980	-.170	.967	-.006	.975	-.061	.977	-.022	.926	-.187	.972	.018
1.074	-.141							.977	-.025		
1.122	-.045										
					LOWER SURFACE						
-.660	.075	-.022	.157	.024	-.085	.025	-.242	.019	-.331	.020	-.317
-.616	-.005	.033	-.033	.075	-.276	0.000	0.000	.066	-.473	.076	-.561
0.000	0.000	.101	-.198	.297	-.232	.130	-.241	.136	-.392	.136	-.482
-.329	-.112	.185	-.203	.400	-.179	.298	-.290	.214	-.212	.221	-.445
-.172	-.062	.398	-.174	.604	-.092	.397	-.202	.292	-.244	.295	-.346
-.030	-.175	.737	.110	.785	.161	.501	-.194	.403	-.260	.396	-.297
.128	-.300			.967	.212	.603	-.008	.489	-.174	.497	-.083
.418	-.150			1.000	.006	.703	.071	.594	-.184	.597	-.014
.564	-.084					.784	.131	.700	.034	.702	.118
.710	.052					.868	.185	.786	.144	.786	.190
.976	.251					.923	.210	.858	.203	.864	.230
1.072	.242							.919	.234	.912	.232
1.110	.181							.967	.153		
0.000	0.000										
CN=	.2479		.2098		.2192		.1733		.1249		.0769
CM=	-.0297		-.0641		-.0780		-.0836		-.0954		-.0916

$\alpha = 1.96^{\circ}$; $C_L = 0.253$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.008	-.021	-.361	.023	-.536	.025	-.541	.022	-.385	.018	-.133
-.567	-.096	.035	-.448	.068	-.725	.079	-.597	.075	-.491	.077	-.349
-.452	-.241	.105	-.379	.134	-.548	.133	-.600	.129	-.430	.129	-.334
-.311	-.332	.178	-.437	.209	-.498	.214	-.473	.201	-.325	.209	-.298
-.023	-.427	.286	-.365	.294	-.369	.295	-.254	.294	-.199	.293	-.290
.133	-.267	.396	-.340	.404	-.328	.407	-.329	.397	-.165	.494	-.334
.272	-.098	.514	-.299	.497	-.327	.502	-.211	.495	-.252	.590	-.376
.416	-.114	.618	-.313	.599	-.394	.601	-.194	.594	-.321	.693	-.182
.565	-.088	.733	-.310	.700	-.391	.698	-.265	.693	-.407	.777	-.140
.713	-.091	.835	-.232	.864	-.186	.863	-.414	.784	-.485	.861	-.169
.854	-.164	.919	-.100	.926	-.107	.923	-.190	.856	-.412	.918	-.131
.980	-.208	.987	.004	.975	-.042	.977	-.047	.926	-.132	.972	.018
1.074	-.195							.977	-.039		
1.122	-.072										
					LOWER SURFACE						
-.660	.082	-.022	.259	.024	.036	.025	-.034	.019	-.072	.020	-.073
-.616	.024	.038	-.006	.075	-.159	0.000	0.000	.066	-.242	.076	-.348
0.000	0.000	.101	-.111	.297	-.183	.130	-.186	.136	-.217	.136	-.373
-.329	-.081	.185	-.150	.400	-.149	.298	-.199	.214	-.204	.221	-.346
-.172	-.049	.398	-.136	.604	-.082	.397	-.166	.292	-.218	.295	-.293
-.030	-.158	.737	.112	.785	.177	.501	-.147	.403	-.098	.396	-.190
.128	-.201			.967	.217	.603	-.010	.489	-.152	.497	-.102
.418	-.115			1.000	.011	.703	.056	.594	-.215	.597	-.022
.564	-.052					.784	.137	.700	.044	.702	.117
.710	.083					.868	.210	.786	.164	.786	.200
.976	.259					.923	.247	.858	.233	.864	.244
1.072	.240							.919	.265	.912	.250
1.110	.172							.967	.158		
0.000	0.000										
CN=	.3217		.3086		.3340		.2879		.2484		.1827
CM=	-.0321		-.0654		-.0794		-.0753		-.0961		-.0859

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Continued.

$\alpha = 2.41^{\circ}$; $C_L = 0.306$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.006	-.021	-.361	.023	-.615	.025	-.612	.022	-.451	.018	-.275
-.567	-.115	.035	-.464	.068	-.750	.079	-.701	.075	-.656	.077	-.384
-.452	-.242	.105	-.404	.134	-.640	.133	-.678	.129	-.562	.129	-.335
-.311	-.347	.178	-.482	.209	-.575	.214	-.608	.201	-.543	.209	-.310
-.023	-.437	.286	-.4C7	.294	-.452	.295	-.255	.294	-.089	.293	-.302
.133	-.329	.396	-.377	.404	-.363	.407	-.333	.397	-.133	.494	-.336
.272	-.114	.514	-.313	.497	-.340	.502	-.370	.495	-.224	.590	-.377
.416	-.102	.618	-.322	.599	-.404	.601	-.400	.594	-.300	.693	-.433
.565	-.097	.733	-.329	.700	-.419	.698	-.190	.693	-.393	.777	-.094
.713	-.104	.835	-.308	.864	-.175	.863	-.347	.784	-.490	.861	-.129
.854	-.170	.919	-.134	.926	-.087	.923	-.191	.856	-.546	.918	-.122
.980	-.219	.987	-.018	.975	-.036	.977	-.039	.926	-.136	.972	.013
1.074	-.219							.977	-.060		
1.122	-.082										
				LOWER SURFACE							
-.660	.093	-.022	.282	.024	.099	.025	.009	.019	.008	.020	-.012
-.616	.044	.038	.025	.075	-.111	0.000	0.000	.066	-.181	.076	-.265
0.000	0.000	.101	-.059	.297	-.164	.130	-.163	.136	-.187	.136	-.311
-.329	-.068	.185	-.109	.400	-.141	.298	-.165	.214	-.168	.221	-.258
-.172	-.037	.398	-.121	.604	-.083	.397	-.158	.292	-.154	.295	-.261
-.030	-.149	.737	.120	.785	.177	.501	-.125	.403	-.090	.396	-.220
.128	-.252			.967	.218	.603	.005	.489	-.139	.497	-.129
.418	-.097			1.000	.003	.703	.071	.594	-.172	.597	-.039
.564	-.038					.784	.154	.700	.045	.702	.109
.710	.093					.868	.231	.786	.169	.786	.196
.976	.268					.923	.269	.858	.242	.864	.241
1.072	.241							.919	.262	.912	.249
1.110	.174							.967	.139		
0.000	0.000										
CN=	.3646		.3583		.3804		.3527		.3023		.2239
CM=	-.0309		-.0718		-.0774		-.0776		-.0961		-.0860

$\alpha = 2.92^{\circ}$; $C_L = 0.369$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.001	-.021	-.650	.023	-.666	.025	-.661	.022	-.542	.018	-.467
-.567	-.120	.035	-.619	.068	-.835	.079	-.752	.075	-.719	.077	-.604
-.452	-.252	.105	-.374	.134	-.742	.133	-.764	.129	-.712	.129	-.647
-.311	-.359	.178	-.472	.209	-.627	.214	-.703	.201	-.691	.209	-.285
-.023	-.448	.286	-.441	.294	-.486	.295	-.629	.294	-.631	.293	-.187
.133	-.338	.396	-.408	.404	-.432	.407	-.306	.397	-.237	.494	-.257
.272	-.129	.514	-.345	.497	-.368	.502	-.375	.495	-.087	.590	-.320
.416	-.096	.618	-.344	.599	-.416	.601	-.417	.594	-.188	.693	-.381
.565	-.104	.733	-.345	.700	-.446	.698	-.405	.693	-.306	.777	-.153
.713	-.120	.835	-.341	.864	-.180	.863	-.250	.784	-.403	.861	-.165
.854	-.186	.919	-.130	.926	-.084	.923	-.157	.856	-.505	.918	-.117
.980	-.235	.987	-.030	.975	-.049	.977	-.022	.926	-.191	.972	.018
1.074	-.257							.977	-.048		
1.122	-.115										
				LOWER SURFACE							
-.660	.095	-.022	.317	.024	.164	.025	.073	.019	.077	.020	.121
-.616	.050	.038	.076	.075	-.069	0.000	0.000	.066	-.147	.076	-.152
0.000	0.000	.101	-.015	.297	-.143	.130	-.135	.136	-.145	.136	-.208
-.329	-.058	.185	-.083	.400	-.128	.298	-.123	.214	-.143	.221	-.192
-.172	-.026	.398	-.098	.604	-.086	.397	-.146	.292	-.123	.295	-.208
-.030	-.137	.737	.120	.785	.178	.501	-.117	.403	-.068	.396	-.150
.128	-.222			.967	.209	.603	.007	.489	-.128	.497	-.116
.418	-.077			1.000	-.019	.703	.074	.594	-.112	.597	-.031
.564	-.023					.784	.164	.700	.068	.702	.105
.710	.104					.868	.247	.786	.182	.786	.186
.976	.273					.923	.292	.858	.251	.864	.236
1.072	.236							.919	.274	.912	.240
1.110	.169							.967	.148		
0.000	0.000										
CN=	.4064		.4170		.4303		.4413		.3829		.2695
CM=	-.0337		-.0710		-.0776		-.0827		-.0863		-.0724

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^0$

(e) M = 0.95. Continued.

$\alpha = 3.42^0$; $C_L = 0.429$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.012	-.021	-.752	.023	-.719	.025	-.714	.022	-.586	.018	-.551
-.567	-.129	.035	-.771	.068	-.870	.079	-.805	.075	-.783	.077	-.743
-.452	-.264	.105	-.258	.134	-.768	.133	-.813	.129	-.762	.129	-.767
-.311	-.368	.178	-.419	.209	-.732	.214	-.790	.201	-.753	.209	-.677
-.023	-.454	.286	-.437	.294	-.488	.295	-.714	.294	-.717	.293	-.660
.133	-.344	.396	-.428	.404	-.501	.407	-.629	.397	-.711	.494	-.016
.272	-.131	.514	-.365	.497	-.392	.502	-.362	.495	-.324	.590	-.121
.416	-.102	.618	-.378	.599	-.429	.601	-.413	.594	-.303	.693	-.170
.565	-.107	.733	-.370	.700	-.474	.698	-.496	.693	-.182	.777	-.286
.713	-.124	.835	-.370	.864	-.177	.863	-.210	.784	-.274	.861	-.201
.854	-.200	.919	-.173	.926	-.092	.923	-.125	.856	-.322	.918	-.156
.980	-.256	.987	-.061	.975	-.067	.977	-.011	.926	-.164	.972	.003
1.074	-.282							.977	-.001		
1.122	-.147										
					LOWER SURFACE						
-.660	.104	-.022	.338	.024	.193	.025	.124	.019	.105	.020	.171
-.616	.063	.038	.113	.075	-.024	0.000	0.000	.066	-.106	.076	-.114
0.000	0.000	.101	.009	.297	-.124	.130	-.114	.136	-.130	.136	-.179
-.329	-.042	.185	-.055	.400	-.123	.298	-.112	.214	-.129	.221	-.170
-.172	-.010	.398	-.078	.604	-.086	.397	-.135	.292	-.103	.295	-.175
-.030	-.123	.737	.120	.785	.176	.501	-.109	.403	-.067	.396	-.138
.128	-.207			.967	.191	.603	.011	.489	-.119	.497	-.107
.418	-.071			1.000	-.039	.703	.076	.594	-.105	.597	-.013
.564	-.010					.784	.159	.700	.082	.702	.113
.710	.114					.868	.246	.786	.191	.786	.196
.976	.277					.923	.298	.858	.264	.864	.241
1.072	.239							.919	.291	.912	.240
1.110	.166							.967	.176		
0.000	0.000										
CN=	.4403		.4526		.4693		.5131		.4632		.3469
CM=	-.0340		-.0773		-.0778		-.0865		-.0831		-.0570

$\alpha = 3.96^0$; $C_L = 0.495$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.019	-.021	-.838	.023	-.789	.025	-.767	.022	-.656	.018	-.620
-.567	-.142	.035	-.935	.068	-.903	.079	-.847	.075	-.822	.077	-.805
-.452	-.276	.105	-.653	.134	-.871	.133	-.856	.129	-.813	.129	-.822
-.311	-.374	.178	-.366	.209	-.795	.214	-.832	.201	-.818	.209	-.790
-.023	-.461	.286	-.446	.294	-.587	.295	-.775	.294	-.757	.293	-.751
.133	-.360	.396	-.439	.404	-.503	.407	-.744	.397	-.735	.494	-.607
.272	-.221	.514	-.382	.497	-.434	.502	-.455	.495	-.695	.590	.016
.416	-.105	.618	-.380	.599	-.444	.601	-.423	.594	-.510	.693	-.005
.565	-.098	.733	-.381	.700	-.503	.698	-.503	.693	-.231	.777	-.145
.713	-.123	.835	-.383	.864	-.193	.863	-.175	.784	-.188	.861	-.185
.854	-.207	.919	-.184	.926	-.102	.923	-.076	.856	-.180	.918	-.163
.980	-.260	.987	-.063	.975	-.078	.977	-.022	.926	-.115	.972	-.019
1.074	-.309							.977	.022		
1.122	-.172										
					LOWER SURFACE						
-.660	.118	-.022	.372	.024	.249	.025	.170	.019	.148	.020	.216
-.616	.088	.038	.146	.075	.026	0.000	0.000	.066	-.076	.076	-.079
0.000	0.000	.101	.047	.297	-.104	.130	-.071	.136	-.100	.136	-.138
-.329	-.021	.185	-.003	.400	-.101	.298	-.103	.214	-.106	.221	-.148
-.172	-.003	.398	-.055	.604	-.084	.397	-.132	.292	-.088	.295	-.152
-.030	-.095	.737	.131	.785	.174	.501	-.104	.403	-.055	.396	-.129
.128	-.173			.967	.191	.603	.015	.489	-.103	.497	-.117
.418	-.052			1.000	-.055	.703	.068	.594	-.088	.597	-.022
.564	.007					.784	.159	.700	.087	.702	.118
.710	.124					.868	.244	.786	.200	.786	.197
.976	.293					.923	.297	.858	.271	.864	.237
1.072	.242							.919	.296	.912	.245
1.110	.163							.967	.193		
0.000	0.000										
CN=	.4939		.5261		.5249		.5578		.5405		.4376
CM=	-.0320		-.0743		-.0801		-.0848		-.0891		-.0584

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(e) M = 0.95. Continued.

$\alpha = 4.92^o$; $C_L = 0.606$

UPPER SURFACE

STA .133		STA .307		STA .480		STA .653		STA .804		STA .933	
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.028	-.021	-.922	.023	-.891	.025	-.859	.022	-.768	.018	-.725
-.567	-.190	.035	-1.083	.066	-.999	.079	-.929	.075	-.909	.077	-.891
-.452	-.299	.105	-.994	.134	-.978	.133	-.939	.129	-.884	.129	-.893
-.311	-.400	.178	-.436	.209	-.925	.214	-.930	.201	-.910	.209	-.877
-.023	-.475	.286	-.387	.294	-.876	.295	-.869	.294	-.853	.293	-.842
.133	-.380	.396	-.470	.404	-.535	.407	-.852	.397	-.836	.494	-.791
.272	-.266	.514	-.408	.497	-.464	.502	-.841	.495	-.814	.590	-.325
.416	-.121	.618	-.409	.599	-.458	.601	-.557	.594	-.786	.693	-.062
.565	-.110	.733	-.407	.700	-.546	.698	-.528	.693	-.382	.777	-.036
.713	-.140	.835	-.411	.864	-.254	.863	-.171	.784	-.260	.861	-.052
.854	-.213	.919	-.212	.926	-.144	.923	-.103	.856	-.194	.918	-.079
.930	-.279	.987	-.082	.975	-.111	.977	-.080	.926	-.124	.972	.012
1.074	-.342							.977	-.099		
1.122	-.209										

LOWER SURFACE

STA .133		STA .307		STA .480		STA .653		STA .804		STA .933	
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.125	-.022	.391	.024	.331	.025	.238	.019	.223	.020	.257
-.616	.101	.038	.189	.075	.096	0.000	0.000	.066	.005	.076	-.015
0.000	0.000	.101	.107	.297	-.052	.130	-.017	.136	-.050	.136	-.096
-.329	.014	.185	.036	.400	-.075	.298	-.066	.214	-.055	.221	-.125
-.172	.018	.398	-.016	.604	-.077	.397	-.103	.292	-.069	.295	-.137
-.030	-.076	.737	.132	.785	.172	.501	-.098	.403	-.044	.396	-.149
.128	-.136			.967	.175	.603	.019	.489	-.099	.497	-.143
.418	-.025			1.000	-.084	.703	.062	.594	-.096	.597	-.070
.564	.033					.784	.149	.700	.068	.702	.085
.710	.139					.868	.244	.786	.179	.786	.164
.976	.299					.923	.302	.858	.240	.864	.223
1.072	.245							.919	.265	.912	.223
1.110	.162							.967	.125		
0.000	0.000										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.5685	.6138	.6299	.6793	.6558	.5071
CM=	-.0269	-.0742	-.0867	-.1009	-.1061	-.0527

$\alpha = 5.95^o$; $C_L = 0.711$

UPPER SURFACE

STA .133		STA .307		STA .480		STA .653		STA .804		STA .933	
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.061	-.021	-1.034	.023	-.999	.025	-.955	.022	-.870	.018	-.832
-.567	-.238	.035	-1.154	.068	-1.099	.079	-1.019	.075	-.999	.077	-.971
-.452	-.324	.105	-1.141	.134	-1.067	.133	-1.022	.129	-.989	.129	-.979
-.311	-.419	.178	-.996	.209	-1.012	.214	-1.004	.201	-.976	.209	-.956
-.023	-.499	.286	-.428	.294	-.975	.295	-.950	.294	-.932	.293	-.909
.133	-.389	.396	-.429	.404	-.913	.407	-.944	.397	-.907	.494	-.814
.272	-.306	.514	-.418	.497	-.607	.502	-.912	.495	-.875	.590	-.446
.416	-.178	.618	-.430	.599	-.513	.601	-.890	.594	-.510	.693	-.135
.565	-.133	.733	-.427	.700	-.581	.698	-.822	.693	-.361	.777	-.065
.713	-.152	.835	-.429	.864	-.242	.863	-.348	.784	-.299	.861	-.037
.854	-.241	.919	-.233	.926	-.164	.923	-.309	.856	-.256	.918	-.035
.980	-.305	.987	-.109	.975	-.139	.977	-.177	.926	-.232	.972	-.002
1.074	-.378							.977	-.226		
1.122	-.253										

LOWER SURFACE

STA .133		STA .307		STA .480		STA .653		STA .804		STA .933	
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.142	-.022	.440	.024	.396	.025	.326	.019	.290	.020	.303
-.616	.136	.038	.254	.075	.162	0.000	0.000	.066	.055	.076	.019
0.000	0.000	.101	.152	.297	.004	.130	.045	.136	-.009	.136	-.078
-.329	.040	.185	.085	.400	-.037	.298	-.028	.214	-.026	.221	-.123
-.172	.051	.398	.019	.604	-.059	.397	-.070	.292	-.040	.295	-.145
-.030	-.038	.737	.150	.785	.175	.501	-.074	.403	-.037	.396	-.171
.128	-.097			.967	.169	.603	.024	.489	-.099	.497	-.180
.418	.015			1.000	-.102	.703	.062	.594	-.138	.597	-.113
.564	.058					.784	.150	.700	.028	.702	.023
.710	.171					.868	.243	.786	.135	.786	.117
.976	.313					.923	.298	.858	.210	.864	.182
1.072	.262							.919	.225	.912	.201
1.110	.163							.967	.060		
C.000	0.000										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.6671	.7268	.7579	.8589	.6839	.5401
CM=	-.0256	-.0743	-.0986	-.1529	-.0969	-.0459

TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Continued.

$\alpha = 7.03^{\circ}$; $C_L = 0.813$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.084	-.021	-1.094	.023	-1.074	.025	-1.023	.022	-.957	.018	-.900
-.567	-.261	.035	-1.227	.068	-1.166	.079	-1.082	.075	-1.058	.077	-1.035
-.452	-.344	.105	-1.179	.134	-1.120	.133	-1.076	.129	-1.043	.129	-1.036
-.311	-.428	.178	-1.121	.209	-1.082	.214	-1.058	.201	-1.036	.209	-.995
-.023	-.507	.286	-.750	.294	-1.045	.295	-1.023	.294	-.987	.293	-.952
.133	-.398	.396	-.604	.404	-1.004	.407	-1.010	.397	-.969	.494	-.790
.272	-.326	.514	-.423	.497	-.796	.502	-.999	.495	-.900	.590	-.499
.416	-.234	.618	-.435	.599	-.676	.601	-.988	.594	-.584	.693	-.347
.565	-.154	.733	-.438	.700	-.601	.698	-.649	.693	-.396	.777	-.237
.713	-.171	.835	-.442	.864	-.305	.863	-.471	.784	-.324	.861	-.160
.854	-.264	.919	-.271	.926	-.232	.923	-.420	.856	-.294	.918	-.156
.980	-.326	.987	-.129	.975	-.190	.977	-.373	.926	-.270	.972	-.104
1.074	-.407							.977	-.261		
1.122	-.289										
					LOWER SURFACE						
-.660	.146	-.022	.455	.024	.442	.025	.377	.019	.335	.020	.342
-.616	.160	.038	.303	.075	.236	0.000	0.000	.066	.116	.076	.055
0.000	0.000	.101	.205	.297	.039	.130	.085	.136	.038	.136	-.045
-.329	.074	.185	.143	.400	-.007	.298	-.009	.214	-.000	.221	-.093
-.172	.081	.398	.061	.604	-.048	.397	-.052	.292	-.021	.295	-.136
-.030	-.003	.737	.158	.785	.169	.501	-.067	.403	-.030	.396	-.171
.128	-.046			.967	.133	.603	.016	.489	-.097	.497	-.196
.418	.044			1.000	-.178	.703	.042	.594	-.162	.597	-.143
.564	.083					.784	.131	.700	.014	.702	-.008
.710	.188					.868	.234	.786	.120	.786	.085
.976	.327					.923	.289	.858	.198	.864	.146
1.072	.257							.919	.213	.912	.159
1.110	.163							.967	.038		
0.000	0.000										
CN=	.7468		.8455		.8603		.9222		.7388		.6092
CM=	-.0261		-.0813		-.1143		-.1638		-.1007		-.0656

$\alpha = 8.15^{\circ}$; $C_L = 0.902$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.115	-.021	-1.163	.023	-1.141	.025	-1.101	.022	-1.034	.018	-.979
-.567	-.302	.035	-1.281	.068	-1.227	.079	-1.155	.075	-1.131	.077	-1.074
-.452	-.380	.105	-1.228	.134	-1.187	.133	-1.145	.129	-1.114	.129	-.976
-.311	-.465	.178	-1.155	.209	-1.150	.214	-1.125	.201	-1.101	.209	-.889
-.023	-.529	.286	-.925	.294	-1.110	.295	-1.093	.294	-1.057	.293	-.774
.133	-.411	.396	-.753	.404	-1.046	.407	-1.079	.397	-1.028	.494	-.555
.272	-.345	.514	-.644	.497	-.855	.502	-1.053	.495	-.972	.590	-.485
.416	-.263	.618	-.520	.599	-.717	.601	-.941	.594	-.691	.693	-.412
.565	-.188	.733	-.455	.700	-.588	.698	-.657	.693	-.500	.777	-.388
.713	-.203	.835	-.462	.864	-.348	.863	-.554	.784	-.401	.861	-.338
.854	-.292	.919	-.325	.926	-.311	.923	-.506	.856	-.373	.918	-.319
.980	-.361	.987	-.161	.975	-.258	.977	-.404	.926	-.353	.972	-.305
1.074	-.440							.977	-.344		
1.122	-.359										
					LOWER SURFACE						
-.660	.163	-.022	.474	.024	.501	.025	.431	.019	.387	.020	.382
-.616	.183	.038	.369	.075	.279	0.000	0.000	.066	.168	.076	.101
0.000	0.000	.101	.261	.297	.082	.130	.125	.136	.072	.136	.002
-.329	.108	.185	.193	.400	.033	.298	.018	.214	.029	.221	-.058
-.172	.108	.398	.096	.604	-.039	.397	-.037	.292	-.008	.295	-.111
-.030	.035	.737	.174	.785	.154	.501	-.071	.403	-.016	.396	-.153
.128	-.005			.967	.100	.603	.006	.489	-.082	.497	-.186
.418	.081			1.000	-.295	.703	.025	.594	-.136	.597	-.160
.564	.120					.784	.112	.700	.017	.702	-.039
.710	.217					.868	.234	.786	.117	.786	.050
.976	.344					.923	.289	.858	.196	.864	.101
1.072	.269							.919	.208	.912	.098
1.110	.168							.967	.024		
0.000	0.000										
CN=	.8513		.9747		.9257		.9816		.8335		.5909
CM=	-.0259		-.1008		-.1197		-.1700		-.1217		-.0758

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) M = 0.95. Continued.

$\alpha = 9.19^{\circ}$; $C_L = 0.969$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
UPPER SURFACE											
-.660	-.172	-.021	-1.223	.023	-1.214	.025	-1.158	.022	-1.113	.018	-.915
-.567	-.361	.035	-1.290	.068	-1.275	.079	-1.204	.075	-1.193	.077	-.870
-.452	-.409	.105	-1.275	.134	-1.233	.133	-1.201	.129	-1.174	.129	-.822
-.311	-.489	.178	-1.154	.209	-1.201	.214	-1.179	.201	-1.156	.209	-.727
-.023	-.541	.286	-1.014	.294	-1.156	.295	-1.128	.294	-1.111	.293	-.683
.133	-.433	.396	-.954	.404	-.997	.407	-1.116	.397	-1.068	.494	-.580
.272	-.360	.514	-.857	.497	-.826	.502	-1.059	.495	-1.009	.590	-.530
.416	-.298	.618	-.798	.599	-.700	.601	-.933	.594	-.772	.693	-.479
.565	-.243	.733	-.612	.700	-.595	.698	-.748	.693	-.617	.777	-.431
.713	-.242	.835	-.529	.864	-.413	.863	-.501	.784	-.550	.861	-.387
.854	-.338	.919	-.342	.926	-.390	.923	-.332	.856	-.504	.918	-.377
.980	-.382	.987	-.193	.975	-.346	.977	-.259	.926	-.490	.972	-.347
1.074	-.471							.977	-.448		
1.122	-.401										
LOWER SURFACE											
-.660	.170	-.022	.495	.024	.529	.025	.482	.019	.422	.020	.424
-.616	.214	.038	.392	.075	.335	0.000	0.000	.066	.207	.076	.161
C.000	0.000	.101	.31C	.297	.115	.130	.178	.136	.112	.136	.035
-.329	.157	.185	.231	.400	.062	.298	.051	.214	.055	.221	-.021
-.172	.134	.398	.130	.604	-.031	.397	-.019	.292	.011	.295	-.075
-.030	.081	.737	.186	.785	.148	.501	-.067	.403	-.003	.396	-.133
.128	.034			.967	.064	.603	.005	.489	-.073	.497	-.178
.418	.119			1.000	-.428	.703	-.004	.594	-.131	.597	-.156
.564	.146					.784	.092	.700	.032	.702	-.048
.710	.251					.868	.218	.786	.136	.786	.029
.976	.357					.923	.267	.858	.211	.864	.085
1.072	.273							.919	.224	.912	.084
1.110	.174							.967	.033		
C.000	0.000										
CN=	.9589		1.1229		.9659		1.0027		.9345		.5794
CM=	-.0241		-.1371		-.1234		-.1541		-.1529		-.0890

$\alpha = 9.58^{\circ}$; $C_L = 0.982$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
UPPER SURFACE											
-.660	-.204	-.021	-1.249	.023	-1.233	.025	-1.212	.022	-1.142	.018	-1.023
-.567	-.374	.035	-1.289	.068	-1.255	.079	-1.216	.075	-1.207	.077	-.821
-.452	-.430	.105	-1.278	.134	-1.219	.133	-1.199	.129	-1.187	.129	-.836
-.311	-.497	.178	-1.146	.209	-1.033	.214	-1.169	.201	-1.149	.209	-.748
-.023	-.545	.286	-1.087	.294	-1.114	.295	-1.153	.294	-1.118	.293	-.758
.133	-.433	.396	-.977	.404	-.868	.407	-1.119	.397	-1.066	.494	-.629
.272	-.369	.514	-.996	.497	-.813	.502	-.583	.495	-.893	.590	-.539
.416	-.307	.618	-.724	.599	-.678	.601	-.684	.594	-.733	.693	-.457
.565	-.232	.733	-.781	.700	-.619	.698	-.453	.693	-.603	.777	-.455
.713	-.269	.835	-.495	.864	-.510	.863	-.425	.784	-.525	.861	-.399
.854	-.343	.919	-.245	.926	-.420	.923	-.358	.856	-.505	.918	-.397
.980	-.398	.987	-.161	.975	-.469	.977	-.303	.926	-.439	.972	-.350
1.074	-.474							.977	-.455		
1.122	-.446										
LOWER SURFACE											
-.660	.165	-.022	.493	.024	.551	.025	.481	.019	.427	.020	.410
-.616	.228	.038	.397	.075	.359	0.000	0.000	.066	.227	.076	.163
0.000	0.000	.101	.326	.297	.104	.130	.181	.136	.098	.136	.025
-.329	.145	.185	.251	.400	.082	.298	.057	.214	.057	.221	-.023
-.172	.145	.398	.140	.604	-.023	.397	-.018	.292	.011	.295	-.077
-.030	.099	.737	.176	.785	.150	.501	-.063	.403	-.010	.396	-.134
.128	.040			.967	.041	.603	-.007	.489	-.091	.497	-.187
.418	.103			1.000	-.427	.703	-.011	.594	-.133	.597	-.162
.564	.171					.784	.065	.700	.018	.702	-.054
.710	.233					.868	.206	.786	.133	.786	.020
.976	.349					.923	.252	.858	.203	.864	.076
1.072	.288							.919	.214	.912	.064
1.110	.168							.967	.033		
0.000	0.000										
CN=	.9768		1.1511		.9571		.8898		.9115		.5981
CM=	-.0209		-.1370		-.1329		-.1105		-.1427		-.0894

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(f) M = 0.97.

$\alpha = -4.99^o$; $C_L = -0.542$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.062	-.021	.371	.023	.363	.025	.330	.022	.427	.018	.466
-.567	.029	.035	.193	.068	.210	.079	.194	.075	.226	.077	.229
-.452	-.070	.105	.117	.134	.127	.133	.145	.129	.170	.129	.162
-.311	-.156	.178	.061	.209	.074	.214	.094	.201	.108	.209	.089
-.023	-.064	.286	.039	.294	.050	.295	.069	.294	.071	.293	.021
.133	-.005	.396	.016	.404	.013	.407	.014	.397	.031	.494	-.114
.272	.039	.514	.003	.497	-.024	.502	-.032	.495	-.028	.590	-.192
.416	.063	.618	-.026	.599	-.065	.601	-.088	.594	-.073	.693	-.299
.565	.085	.733	-.063	.700	-.109	.698	-.168	.693	-.163	.777	-.400
.713	.089	.835	-.115	.864	-.221	.863	-.331	.784	-.263	.861	-.507
.854	.036	.919	-.103	.926	-.196	.923	-.218	.856	-.389	.918	-.654
.980	-.018	.987	.005	.975	-.060	.977	-.082	.926	-.570	.972	-.306
1.074	-.080							.977	-.578		
1.122	-.055										
					LOWER SURFACE						
-.660	-.035	-.022	-.577	.024	-.931	.025	-1.001	.019	-.947	.020	-.454
-.616	-.152	.038	-.842	.075	-.967	0.000	0.000	.066	-.953	.076	-.421
0.000	0.000	.101	-.748	.297	-.948	.130	-1.113	.136	-.932	.136	-.419
-.329	-.270	.185	-.634	.400	-.942	.298	-1.046	.214	-.754	.221	-.399
-.172	-.087	.398	-.579	.604	-.315	.397	-.785	.292	-.755	.295	-.376
-.030	-.206	.737	-.170	.785	.038	.501	-.384	.403	-.684	.396	-.355
.128	-.331			.967	.035	.603	-.325	.489	-.624	.497	-.336
.418	-.282			1.000	-.002	.703	-.059	.594	-.558	.597	-.301
.564	-.353					.784	.072	.700	-.462	.702	-.291
.710	-.248					.868	.022	.786	-.452	.786	-.276
.976	.145					.923	-.097	.858	-.393	.864	-.299
1.072	.194							.919	-.390	.912	-.284
1.110	.163							.967	-.397		
0.000	0.000										
CN= -.2672		-.5195		-.5140		-.4901		-.5566		-.2115	
CM= -.0068		.0011		-.0289		-.0483		.0141		-.0386	

$\alpha = -3.98^o$; $C_L = -0.455$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.069	-.021	.330	.023	.286	.025	.279	.022	.381	.018	.443
-.567	.025	.035	.134	.068	.141	.079	.139	.075	.179	.077	.196
-.452	-.095	.105	.072	.134	.071	.133	.081	.129	.129	.129	.135
-.311	-.186	.178	.021	.209	.017	.214	.044	.201	.074	.209	.065
-.023	-.110	.286	-.010	.294	-.008	.295	.025	.294	.036	.293	.009
.133	-.021	.396	-.035	.404	-.029	.407	-.019	.397	.009	.494	-.115
.272	.010	.514	-.036	.497	-.059	.502	-.063	.495	-.036	.590	-.189
.416	.042	.618	-.060	.599	-.090	.601	-.106	.594	-.086	.693	-.283
.565	.068	.733	-.090	.700	-.156	.698	-.176	.693	-.166	.777	-.387
.713	.062	.835	-.135	.864	-.211	.863	-.232	.784	-.252	.861	-.499
.854	.007	.919	-.108	.926	-.156	.923	-.178	.856	-.339	.918	-.651
.980	-.035	.937	.000	.975	-.017	.977	-.033	.926	-.483	.972	-.315
1.074	-.092							.977	-.545		
1.122	-.053										
					LOWER SURFACE						
-.660	-.007	-.022	-.492	.024	-.854	.025	-.927	.019	-1.018	.020	-.495
-.615	-.112	.038	-.745	.075	-.882	0.000	0.000	.066	-1.064	.076	-.496
0.000	0.000	.101	-.613	.297	-.876	.130	-1.040	.136	-1.030	.136	-.416
-.329	-.250	.185	-.579	.400	-.888	.298	-.989	.214	-.929	.221	-.381
-.172	-.094	.398	-.554	.604	-.197	.397	-.936	.292	-.804	.295	-.348
-.030	-.225	.737	-.096	.785	.039	.501	-.390	.403	-.660	.396	-.315
.128	-.310			.967	.078	.603	-.167	.489	-.613	.497	-.288
.418	-.277			1.000	.020	.703	.041	.594	-.571	.597	-.282
.564	-.329					.784	.170	.700	-.491	.702	-.286
.710	-.169					.868	.178	.786	-.368	.786	-.284
.976	.159					.923	.113	.858	-.284	.864	-.249
1.072	.204							.919	-.173	.912	-.252
1.110	.155							.967	-.093		
0.000	0.000										
CN= -.1955		-.4122		-.4174		-.4161		-.5423		-.1912	
CM= -.0146		-.0155		-.0396		-.0626		-.0087		-.0442	

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = -2.98^{\circ}$; $C_L = -0.352$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	.067	-.021	.266	.023	.206	.025	.191	.022	.322	.018	.415
-.567	.008	.035	.054	.068	.045	.079	.066	.075	.132	.077	.157
-.452	-.106	.105	.003	.134	-.001	.133	.030	.129	.076	.129	.079
-.311	-.207	.179	-.043	.209	-.048	.214	.003	.201	.023	.209	.033
-.023	-.269	.286	-.061	.294	-.050	.295	-.011	.294	.003	.293	-.032
.133	-.032	.396	-.075	.404	-.075	.407	-.042	.397	-.022	.494	-.142
.272	-.010	.514	-.070	.497	-.099	.502	-.084	.495	-.083	.590	-.211
.416	.021	.618	-.090	.599	-.112	.601	-.133	.594	-.095	.693	-.299
.565	.047	.733	-.120	.700	-.181	.698	-.200	.693	-.161	.777	-.372
.713	.044	.835	-.155	.864	-.200	.863	-.254	.784	-.273	.861	-.277
.854	-.014	.919	-.111	.926	-.131	.923	-.169	.856	-.357	.918	-.451
.980	-.056	.987	.005	.975	-.008	.977	-.018	.926	-.215	.972	-.399
1.074	-.093							.977	-.105		
1.122	-.033										
				LOWER SURFACE							
-.660	.011	-.022	-.336	.024	-.740	.025	-.831	.019	-.917	.020	-.580
-.616	-.080	.038	-.546	.075	-.755	0.000	0.000	.066	-1.056	.076	-.569
0.000	0.000	.101	-.528	.297	-.772	.130	-.982	.136	-.970	.136	-.477
-.329	-.215	.185	-.442	.400	-.698	.298	-.957	.214	-.962	.221	-.428
-.172	-.089	.398	-.509	.604	-.174	.397	-.898	.292	-.802	.295	-.425
-.030	-.228	.737	-.060	.785	.084	.501	-.243	.403	-.635	.396	-.323
.128	-.358			.967	.107	.603	-.034	.489	-.590	.497	-.314
.418	-.260			1.000	.041	.703	.123	.594	-.481	.597	-.281
.564	-.290					.784	.202	.700	-.305	.702	-.273
.710	-.104					.868	.226	.786	-.241	.786	-.262
.976	.184					.923	.196	.858	-.088	.864	-.252
1.072	.204							.919	-.019	.912	-.228
1.110	.171							.967	.114		
0.000	0.000										
CN=	-.1099		-.2896		-.2981		-.3163		-.4707		-.2123
CM=	-.0138		-.0203		-.0486		-.0808		-.0221		-.0358

$\alpha = -2.04^{\circ}$; $C_L = -0.241$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
				UPPER SURFACE							
-.660	.063	-.021	.190	.023	.130	.025	.124	.022	.269	.018	.351
-.567	-.002	.035	-.024	.068	-.034	.079	.010	.075	.056	.077	.095
-.452	-.134	.105	-.068	.134	-.066	.133	-.015	.129	.032	.129	.029
-.311	-.228	.178	-.106	.209	-.100	.214	-.051	.201	-.018	.209	-.018
-.023	-.318	.286	-.124	.294	-.096	.295	-.060	.294	-.035	.293	-.069
.133	-.094	.396	-.127	.404	-.108	.407	-.083	.397	-.059	.494	-.156
.272	-.033	.514	-.101	.497	-.150	.502	-.119	.495	-.129	.590	-.232
.416	-.016	.618	-.130	.599	-.162	.601	-.167	.594	-.182	.693	-.328
.565	.018	.733	-.146	.700	-.187	.698	-.219	.693	-.252	.777	-.372
.713	.015	.835	-.170	.864	-.231	.863	-.332	.784	-.263	.861	-.128
.854	-.039	.919	-.118	.926	-.132	.923	-.178	.856	-.343	.918	-.195
.980	-.074	.987	.005	.975	-.035	.977	-.019	.926	-.178	.972	-.243
1.074	-.101							.977	-.008		
1.122	-.037										
				LOWER SURFACE							
-.660	.026	-.022	-.180	.024	-.606	.025	-.764	.019	-.809	.020	-.728
-.616	-.050	.038	-.422	.075	-.646	0.000	0.000	.066	-.977	.076	-.685
0.000	0.000	.101	-.487	.297	-.678	.130	-.901	.136	-.935	.136	-.640
-.329	-.173	.185	-.322	.400	-.562	.298	-.846	.214	-.901	.221	-.500
-.172	-.085	.398	-.472	.604	-.152	.397	-.509	.292	-.817	.295	-.428
-.030	-.218	.737	.039	.785	.137	.501	-.183	.403	-.529	.396	-.394
.128	-.339			.967	.206	.603	-.007	.489	-.425	.497	-.309
.418	-.252			1.000	.027	.703	.095	.594	-.250	.597	-.261
.564	-.227					.784	.143	.700	-.081	.702	-.232
.710	-.069					.868	.211	.786	.018	.786	-.198
.976	.201					.923	.218	.858	.123	.864	-.129
1.072	.219							.919	.178	.912	-.135
1.110	.179							.967	.143		
0.000	0.000										
CN=	-.0209		-.1495		-.1750		-.2025		-.3031		-.2283
CM=	-.0153		-.0402		-.0683		-.0917		-.0719		-.0348

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TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = -1.06^{\circ}$; $C_L = -0.114$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.066	-.021	.096	.023	-.000	.025	.027	.022	.187	.018	.294
-.567	-.016	.035	-.074	.068	-.143	.079	-.083	.075	-.005	.077	.015
-.452	-.142	.105	-.138	.134	-.142	.133	-.112	.129	-.041	.129	-.021
-.311	-.239	.178	-.173	.209	-.160	.214	-.096	.201	-.067	.209	-.069
-.023	-.339	.286	-.164	.294	-.148	.295	-.101	.294	-.073	.293	-.111
.133	-.228	.396	-.166	.404	-.140	.407	-.110	.397	-.094	.494	-.204
.272	-.040	.514	-.139	.497	-.179	.502	-.138	.495	-.153	.590	-.259
.416	-.023	.618	-.161	.599	-.200	.601	-.171	.594	-.216	.693	-.343
.565	-.001	.733	-.166	.700	-.206	.698	-.259	.693	-.291	.777	-.417
.713	-.003	.835	-.187	.864	-.217	.863	-.418	.784	-.355	.861	-.067
.854	-.057	.919	-.108	.926	-.147	.923	-.166	.856	-.312	.918	-.011
.980	-.090	.987	.013	.975	-.041	.977	-.014	.926	-.206	.972	.059
1.074	-.102							.977	-.031		
1.122	-.018										
					LOWER SURFACE						
-.660	.052	-.022	-.070	.024	-.428	.025	-.620	.019	-.677	.020	-.760
-.616	-.040	.038	-.317	.075	-.550	0.000	0.000	.066	-.880	.076	-.845
0.000	0.000	.101	-.372	.297	-.572	.130	-.811	.136	-.821	.136	-.804
-.329	-.136	.185	-.351	.400	-.202	.298	-.365	.214	-.824	.221	-.783
-.172	-.071	.398	-.350	.604	-.180	.397	-.284	.292	-.611	.295	-.644
-.030	-.201	.737	.075	.785	.144	.501	-.261	.403	-.290	.396	-.531
.128	-.334			.967	.220	.603	-.022	.489	-.216	.497	-.307
.418	-.248			1.000	.019	.703	.069	.594	-.171	.597	-.206
.564	-.161					.784	.127	.700	-.010	.702	-.053
.710	-.002					.868	.188	.786	.096	.786	.105
.976	.229					.923	.198	.858	.166	.864	.176
1.072	.237							.919	.212	.912	.232
1.110	.185							.967	.140		
0.000	0.000										
CN=	.0630		-.0479		-.0445		-.0646		-.1478		-.1943
CM=	-.0214		-.0504		-.0736		-.0942		-.0961		-.0798

$\alpha = -0.07^{\circ}$; $C_L = 0.010$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.056	-.021	-.037	.023	-.154	.025	-.100	.022	.062	.018	.190
-.567	-.041	.035	-.164	.068	-.261	.079	-.205	.075	-.117	.077	-.060
-.452	-.172	.105	-.188	.134	-.243	.133	-.216	.129	-.136	.129	-.100
-.311	-.275	.178	-.240	.209	-.266	.214	-.206	.201	-.123	.209	-.125
-.023	-.358	.286	-.213	.294	-.234	.295	-.177	.294	-.115	.293	-.169
.133	-.282	.396	-.218	.404	-.189	.407	-.168	.397	-.130	.494	-.258
.272	-.150	.514	-.185	.497	-.231	.502	-.183	.495	-.191	.590	-.324
.416	-.034	.618	-.209	.599	-.250	.601	-.187	.594	-.250	.693	-.393
.565	-.022	.733	-.210	.700	-.313	.698	-.255	.693	-.334	.777	-.449
.713	-.027	.835	-.223	.864	-.214	.863	-.477	.784	-.414	.861	-.045
.854	-.079	.919	-.108	.926	-.128	.923	-.173	.856	-.427	.918	.002
.980	-.121	.987	.008	.975	-.036	.977	-.022	.926	-.187	.972	.066
1.074	-.125							.977	-.037		
1.122	-.036										
					LOWER SURFACE						
-.660	.067	-.022	.045	.024	-.277	.025	-.478	.019	-.533	.020	-.571
-.616	-.011	.038	-.229	.075	-.429	0.000	0.000	.066	-.750	.076	-.749
0.000	0.000	.101	-.296	.297	-.319	.130	-.544	.136	-.669	.136	-.748
-.329	-.118	.185	-.293	.400	-.224	.298	-.349	.214	-.561	.221	-.761
-.172	-.061	.398	-.260	.604	-.195	.397	-.282	.292	-.249	.295	-.394
-.030	-.187	.737	.095	.785	.153	.501	-.244	.403	-.270	.396	-.372
.128	-.311			.967	.225	.603	-.019	.489	-.188	.497	-.156
.418	-.227			1.000	.022	.703	.073	.594	-.253	.597	-.006
.564	-.092					.784	.120	.700	.002	.702	.106
.710	.043					.868	.180	.786	.103	.786	.161
.976	.247					.923	.188	.858	.180	.864	.196
1.072	.242							.919	.212	.912	.204
1.110	.186							.967	.135		
0.000	0.000										
CN=	.1635		.0653		.0823		.0414		-.0162		-.0359
CM=	-.0250		-.0590		-.0763		-.0908		-.0999		-.1020

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = 0.97^{\circ}$; $C_L = 0.132$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.047	-.021	-.273	.023	-.323	.025	-.275	.022	-.125	.018	.073
-.567	-.052	.035	-.472	.068	-.439	.079	-.378	.075	-.281	.077	-.164
-.452	-.182	.105	-.268	.134	-.389	.133	-.339	.129	-.265	.129	-.163
-.311	-.289	.178	-.263	.209	-.352	.214	-.273	.201	-.242	.209	-.179
-.023	-.391	.286	-.261	.294	-.328	.295	-.252	.294	-.204	.293	-.209
.133	-.305	.396	-.265	.404	-.244	.407	-.253	.397	-.149	.494	-.281
.272	-.231	.514	-.226	.497	-.262	.502	-.286	.495	-.161	.590	-.339
.416	-.056	.618	-.245	.599	-.318	.601	-.259	.594	-.234	.693	-.421
.565	-.022	.733	-.245	.700	-.349	.698	-.213	.693	-.330	.777	-.501
.713	-.035	.835	-.269	.864	-.194	.863	-.422	.784	-.421	.861	-.043
.854	-.106	.919	-.115	.926	-.105	.923	-.169	.856	-.553	.918	.004
.980	-.155	.987	.003	.975	-.032	.977	-.018	.926	-.188	.972	.052
1.074	-.164							.977	-.066		
1.122	-.053										
					LOWER SURFACE						
-.660	.083	-.022	.103	.024	-.054	.025	-.281	.019	-.326	.020	-.282
-.616	.016	.038	-.082	.075	-.275	0.000	0.000	.066	-.519	.076	-.483
0.000	0.000	.101	-.131	.297	-.264	.130	-.269	.136	-.431	.136	-.452
-.329	-.098	.185	-.188	.400	-.223	.298	-.225	.214	-.371	.221	-.399
-.172	-.047	.398	-.169	.604	-.170	.397	-.284	.292	-.123	.295	-.368
-.030	-.172	.737	.105	.785	.164	.501	-.264	.403	-.218	.396	-.395
.128	-.294			.967	.213	.603	-.005	.489	-.146	.497	-.178
.418	-.198			1.000	.003	.703	.081	.594	-.249	.597	-.008
.564	-.069					.784	.131	.700	.022	.702	.102
.710	.067					.868	.188	.786	.141	.786	.162
.976	.263					.923	.207	.858	.207	.864	.190
1.072	.248							.919	.234	.912	.198
1.110	.181							.967	.133		
0.000	0.000										
CN=	.2374		.2071		.1888		.1664		.1204		.0779
CM=	-.0293		-.0596		-.0717		-.0807		-.1007		-.0937

$\alpha = 1.96^{\circ}$; $C_L = 0.249$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	.035	-.021	-.458	.023	-.498	.025	-.468	.022	-.290	.018	-.184
-.567	-.072	.035	-.629	.068	-.654	.079	-.549	.075	-.488	.077	-.402
-.452	-.201	.105	-.415	.134	-.556	.133	-.480	.129	-.410	.129	-.381
-.311	-.298	.178	-.433	.209	-.379	.214	-.411	.201	-.377	.209	-.303
-.023	-.403	.286	-.206	.294	-.374	.295	-.337	.294	-.329	.293	-.267
.133	-.325	.396	-.263	.404	-.319	.407	-.311	.397	-.278	.494	-.267
.272	-.278	.514	-.246	.497	-.305	.502	-.346	.495	-.339	.590	-.265
.416	-.173	.618	-.265	.599	-.368	.601	-.373	.594	-.372	.693	-.327
.565	-.025	.733	-.273	.700	-.385	.698	-.434	.693	-.353	.777	-.451
.713	-.030	.835	-.296	.864	-.210	.863	-.219	.784	-.295	.861	-.067
.854	-.112	.919	-.161	.926	-.095	.923	-.107	.856	-.366	.918	-.050
.980	-.165	.987	-.016	.975	-.060	.977	-.024	.926	-.224	.972	.044
1.074	-.191							.977	-.045		
1.122	-.087										
					LOWER SURFACE						
-.660	.100	-.022	.217	.024	.084	.025	-.053	.019	-.146	.020	-.065
-.616	.047	.038	-.007	.075	-.152	0.000	0.000	.066	-.280	.076	-.287
0.000	0.000	.101	-.090	.297	-.191	.130	-.185	.136	-.234	.136	-.329
-.329	-.072	.185	-.126	.400	-.166	.298	-.233	.214	-.212	.221	-.324
-.172	-.032	.398	-.122	.604	-.135	.397	-.278	.292	-.211	.295	-.333
-.030	-.145	.737	.117	.785	.164	.501	-.185	.403	-.232	.396	-.253
.128	-.274			.967	.210	.603	.012	.489	-.116	.497	-.182
.418	-.116			1.000	-.016	.703	.069	.594	-.215	.597	-.014
.564	-.050					.784	.144	.700	.048	.702	.117
.710	.081					.868	.212	.786	.163	.786	.200
.976	.269					.923	.237	.858	.234	.864	.234
1.072	.251							.919	.255	.912	.244
1.110	.181							.967	.151		
0.000	0.000										
CN=	.3205		.3034		.3019		.2820		.2494		.1845
CM=	-.0298		-.0606		-.0740		-.0785		-.0943		-.0848

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = 2.43^{\circ}$; $C_L = 0.304$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	.028	-.021	-.541	.023	-.591	.025	-.558	.022	-.416	.018	-.314
	-.567	-.087	.035	-.710	.068	-.728	.079	-.625	.075	-.580	.077	-.546
	-.452	-.216	.105	-.555	.134	-.644	.133	-.646	.129	-.583	.129	-.503
	-.311	-.320	.178	-.458	.209	-.626	.214	-.540	.201	-.471	.209	-.410
	-.023	-.414	.286	-.238	.294	-.348	.295	-.271	.294	-.304	.293	-.363
	.133	-.341	.396	-.272	.404	-.331	.407	-.333	.397	-.310	.494	-.258
	.272	-.288	.514	-.256	.497	-.315	.502	-.361	.495	-.358	.590	-.269
	.416	-.196	.618	-.279	.599	-.371	.601	-.390	.594	-.395	.693	-.245
	.565	-.035	.733	-.277	.700	-.402	.698	-.455	.693	-.471	.777	-.176
	.713	-.039	.835	-.309	.864	-.246	.863	-.202	.784	-.314	.861	-.139
	.854	-.113	.919	-.167	.926	-.106	.923	-.104	.856	-.208	.918	-.121
	.980	-.172	.987	-.025	.975	-.073	.977	-.036	.926	-.149	.972	.033
	1.074	-.211							.977	-.014		
	1.122	-.098										
						LOWER SURFACE						
	-.660	.111	-.022	.267	.024	.120	.025	.015	.019	-.038	.020	.050
	-.616	.053	.038	.034	.075	-.086	0.000	0.000	.066	-.251	.076	-.229
	0.000	0.000	.101	-.066	.297	-.157	.130	-.157	.136	-.192	.136	-.304
	-.329	-.056	.185	-.106	.400	-.134	.298	-.231	.214	-.190	.221	-.274
	-.172	-.019	.398	-.099	.604	-.064	.397	-.229	.292	-.194	.295	-.233
	-.030	-.136	.737	.121	.785	.165	.501	-.182	.403	-.066	.396	-.220
	.128	-.263			.967	.201	.603	.010	.489	-.119	.497	-.146
	.418	-.099			1.000	-.039	.703	.057	.594	-.185	.597	-.028
	.564	-.033					.784	.146	.700	.064	.702	.115
	.710	.099					.868	.226	.786	.177	.786	.194
	.976	.278					.923	.264	.858	.246	.864	.239
	1.072	.252							.919	.269	.912	.247
	1.110	.180							.967	.162		
	0.000	0.000										
CN=	.3643		.3533		.3783		.3285		.3166		.2356	
CM=	-.0282		-.0596		-.0791		-.0767		-.0914		-.0708	

$\alpha = 2.93^{\circ}$; $C_L = 0.366$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	.018	-.021	-.674	.023	-.647	.025	-.606	.022	-.461	.018	-.399
	-.567	-.131	.035	-.749	.068	-.805	.079	-.735	.075	-.676	.077	-.628
	-.452	-.226	.105	-.636	.134	-.735	.133	-.713	.129	-.678	.129	-.664
	-.311	-.327	.178	-.496	.209	-.674	.214	-.618	.201	-.594	.209	-.590
	-.023	-.430	.286	-.409	.294	-.312	.295	-.564	.294	-.582	.293	-.555
	.133	-.345	.396	-.271	.404	-.323	.407	-.296	.397	-.256	.494	-.337
	.272	-.294	.514	-.264	.497	-.325	.502	-.346	.495	-.333	.590	-.398
	.416	-.207	.618	-.290	.599	-.371	.601	-.390	.594	-.388	.693	-.002
	.565	-.052	.733	-.286	.700	-.418	.698	-.462	.693	-.473	.777	-.117
	.713	-.061	.835	-.319	.864	-.294	.863	-.258	.784	-.552	.861	-.180
	.854	-.122	.919	-.177	.926	-.124	.923	-.111	.856	-.199	.918	-.131
	.980	-.186	.987	-.035	.975	-.086	.977	-.062	.926	-.086	.972	.031
	1.074	-.230							.977	-.012		
	1.122	-.124										
						LOWER SURFACE						
	-.660	.113	-.022	.284	.024	.166	.025	.076	.019	.017	.020	.093
	-.616	.079	.038	.057	.075	-.065	0.000	0.000	.066	-.192	.076	-.183
	0.000	0.000	.101	-.030	.297	-.131	.130	-.131	.136	-.178	.136	-.257
	-.329	-.045	.185	-.078	.400	-.113	.298	-.217	.214	-.172	.221	-.242
	-.172	-.011	.398	-.082	.604	-.069	.397	-.115	.292	-.195	.295	-.229
	-.030	-.125	.737	.125	.785	.165	.501	-.206	.403	-.045	.396	-.189
	.128	-.256			.967	.191	.603	.003	.489	-.123	.497	-.149
	.418	-.088			1.000	-.060	.703	.051	.594	-.188	.597	-.022
	.564	-.024					.784	.136	.700	.069	.702	.129
	.710	.108					.868	.219	.786	.182	.786	.205
	.976	.277					.923	.267	.858	.249	.864	.246
	1.072	.256							.919	.278	.912	.247
	1.110	.178							.967	.172		
	0.000	0.000										
CN=	.4026		.4121		.4113		.3964		.3852		.3145	
CM=	-.0261		-.0598		-.0806		-.0787		-.0958		-.0656	

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = 3.44^{\circ}$; $C_L = 0.429$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.010	-.021	-.724	.023	-.706	.025	-.681	.022	-.530	.018	-.479
-.567	-.127	.035	-.786	.068	-.855	.079	-.756	.075	-.723	.077	-.683
-.452	-.240	.105	-.744	.134	-.804	.133	-.767	.129	-.718	.129	-.711
-.311	-.340	.178	-.538	.209	-.752	.214	-.762	.201	-.727	.209	-.678
-.023	-.444	.286	-.469	.294	-.714	.295	-.673	.294	-.684	.293	-.653
.133	-.351	.396	-.288	.404	-.288	.407	-.476	.397	-.681	.494	-.590
.272	-.313	.514	-.271	.497	-.321	.502	-.324	.495	-.316	.590	-.337
.416	-.220	.613	-.298	.599	-.376	.601	-.381	.594	-.367	.693	-.006
.565	-.077	.733	-.309	.700	-.440	.698	-.459	.693	-.460	.777	-.060
.713	-.067	.835	-.328	.864	-.334	.863	-.300	.784	-.556	.861	-.110
.854	-.140	.919	-.188	.926	-.149	.923	-.129	.856	-.189	.918	-.117
.980	-.197	.987	-.047	.975	-.101	.977	-.080	.926	-.071	.972	.031
1.074	-.247							.977	-.021		
1.122	-.141										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.125	-.022	.317	.024	.230	.025	.130	.019	.074	.020	.132
-.616	.082	.038	.087	.075	-.005	0.000	0.000	.066	-.163	.076	-.157
0.000	0.000	.101	.013	.297	-.118	.130	-.099	.136	-.155	.136	-.229
-.329	-.036	.165	-.035	.400	-.109	.298	-.199	.214	-.149	.221	-.223
-.172	-.002	.398	-.070	.604	-.070	.397	-.104	.292	-.176	.295	-.200
-.030	-.119	.737	.128	.785	.163	.501	-.206	.403	-.045	.396	-.190
.128	-.240			.967	.182	.603	.008	.489	-.119	.497	-.160
.418	-.078			1.000	-.070	.703	.043	.594	-.203	.597	-.038
.564	-.010					.784	.127	.700	.063	.702	.115
.710	.118					.868	.217	.786	.175	.786	.191
.976	.288					.923	.268	.858	.250	.864	.240
1.072	.254							.919	.277	.912	.246
1.110	.177							.967	.169		
0.000	0.000										

	.133	.307	.480	.653	.804	.933
CN=	.4432	.4631	.4857	.4603	.4583	.3709
CM=	-.0284	-.0604	-.0827	-.0817	-.0961	-.0622

$\alpha = 3.95^{\circ}$; $C_L = 0.493$

UPPER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	-.010	-.021	-.770	.023	-.766	.025	-.722	.022	-.586	.018	-.544
-.567	-.140	.035	-.897	.068	-.901	.079	-.810	.075	-.770	.077	-.741
-.452	-.240	.105	-.812	.134	-.851	.133	-.814	.129	-.754	.129	-.757
-.311	-.347	.178	-.561	.209	-.833	.214	-.803	.201	-.768	.209	-.733
-.023	-.448	.285	-.489	.294	-.744	.295	-.754	.294	-.722	.293	-.702
.133	-.365	.396	-.329	.404	-.389	.407	-.734	.397	-.721	.494	-.674
.272	-.322	.514	-.278	.497	-.312	.502	-.429	.495	-.674	.590	-.658
.416	-.247	.618	-.305	.599	-.374	.601	-.376	.594	-.425	.693	-.131
.565	-.080	.733	-.311	.700	-.448	.698	-.458	.693	-.464	.777	-.033
.713	-.083	.835	-.335	.864	-.377	.863	-.267	.784	-.467	.861	-.015
.854	-.151	.919	-.185	.926	-.158	.923	-.142	.856	-.179	.918	.006
.980	-.208	.987	-.061	.975	-.106	.977	-.111	.926	-.080	.972	.061
1.074	-.259							.977	-.047		
1.122	-.156										

LOWER SURFACE

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
-.660	.122	-.022	.348	.024	.259	.025	.164	.019	.122	.020	.158
-.616	.093	.038	.127	.075	.029	0.000	0.000	.066	-.107	.076	-.117
0.000	0.000	.101	.046	.297	-.088	.130	-.079	.136	-.133	.136	-.208
-.329	-.012	.185	-.001	.400	-.096	.298	-.162	.214	-.141	.221	-.196
-.172	.008	.398	-.046	.604	-.070	.397	-.127	.292	-.161	.295	-.187
-.030	-.103	.737	.134	.785	.168	.501	-.204	.403	-.058	.396	-.190
.128	-.211			.967	.181	.603	.008	.489	-.116	.497	-.196
.418	-.062			1.000	-.077	.703	.040	.594	-.212	.597	-.069
.564	.004					.784	.120	.700	.053	.702	.085
.710	.132					.868	.219	.786	.169	.786	.178
.976	.292					.923	.271	.858	.239	.864	.225
1.072	.254							.919	.267	.912	.232
1.110	.182							.967	.149		
0.000	0.000										

	.133	.307	.480	.653	.804	.933
CN=	.4843	.5116	.5329	.5219	.5153	.4255
CM=	-.0289	-.0602	-.0861	-.0852	-.0991	-.0624

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{o}$

(f) M = 0.97. Continued.

$\alpha = 4.98^{o}$; $C_L = 0.609$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.020	-.021	-.888	.023	-.864	.025	-.810	.022	-.711	.018	-.646
-.567	-.183	.035	-1.026	.068	-.966	.079	-.897	.075	-.859	.077	-.822
-.452	-.271	.105	-.560	.134	-.944	.133	-.897	.129	-.847	.129	-.838
-.311	-.367	.178	-.830	.209	-.923	.214	-.886	.201	-.859	.209	-.812
-.023	-.466	.286	-.500	.294	-.857	.295	-.834	.294	-.803	.293	-.782
.133	-.370	.396	-.489	.404	-.803	.407	-.822	.397	-.798	.494	-.761
.272	-.330	.514	-.309	.497	-.494	.502	-.807	.495	-.775	.590	-.739
.416	-.251	.618	-.321	.599	-.367	.601	-.800	.594	-.766	.693	-.238
.565	-.112	.733	-.332	.700	-.472	.698	-.505	.693	-.562	.777	-.173
.713	-.100	.835	-.348	.864	-.372	.863	-.261	.784	-.299	.861	-.096
.854	-.169	.919	-.200	.926	-.175	.923	-.175	.856	-.277	.918	-.058
.980	-.229	.987	-.074	.975	-.117	.977	-.160	.926	-.268	.972	-.037
1.074	-.293							.977	-.266		
1.122	-.197										
				LOWER SURFACE							
-.660	.143	-.022	.396	.024	.339	.025	.248	.019	.214	.020	.223
-.616	.122	.038	.197	.075	.108	0.000	0.000	.066	-.027	.076	-.069
0.000	0.000	.101	.111	.297	-.043	.130	-.010	.136	-.067	.136	-.160
-.329	.014	.185	.044	.400	-.073	.298	-.071	.214	-.104	.221	-.170
-.172	.032	.398	-.007	.604	-.064	.397	-.113	.292	-.113	.295	-.174
-.030	-.070	.737	.146	.785	.179	.501	-.157	.403	-.059	.396	-.210
.128	-.152			.967	.176	.603	.007	.489	-.112	.497	-.245
.418	-.020			1.000	-.090	.703	.038	.594	-.225	.597	-.165
.564	.038					.784	.118	.700	.024	.702	-.002
.710	.152					.868	.219	.786	.142	.786	.092
.976	.313					.923	.276	.858	.208	.864	.150
1.072	.267							.919	.220	.912	.154
1.110	.177							.967	.061		
0.000	0.000										
CN=	.5710		.6269		.6571		.6827		.6288		.4827
CM=	-.0277		-.0625		-.0959		-.1129		-.1162		-.0637

$\alpha = 5.96^{o}$; $C_L = 0.709$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.052	-.021	-.965	.023	-.939	.025	-.902	.022	-.803	.018	-.729
-.567	-.219	.035	-1.104	.068	-1.047	.079	-.962	.075	-.930	.077	-.894
-.452	-.293	.105	-1.071	.134	-1.025	.133	-.969	.129	-.916	.129	-.906
-.311	-.387	.178	-1.012	.209	-.993	.214	-.959	.201	-.923	.209	-.885
-.023	-.488	.286	-.570	.294	-.953	.295	-.900	.294	-.875	.293	-.853
.133	-.395	.396	-.528	.404	-.930	.407	-.908	.397	-.859	.494	-.836
.272	-.339	.514	-.369	.497	-.694	.502	-.883	.495	-.852	.590	-.768
.416	-.276	.618	-.340	.599	-.471	.601	-.859	.594	-.830	.693	-.369
.565	-.156	.733	-.358	.700	-.532	.698	-.884	.693	-.393	.777	-.279
.713	-.131	.835	-.355	.864	-.314	.863	-.455	.784	-.343	.861	-.194
.854	-.196	.919	-.205	.926	-.202	.923	-.397	.856	-.315	.918	-.148
.980	-.255	.987	-.095	.975	-.161	.977	-.334	.926	-.304	.972	-.115
1.074	-.337							.977	-.290		
1.122	-.237										
				LOWER SURFACE							
-.660	.151	-.022	.425	.024	.405	.025	.320	.019	.268	.020	.271
-.616	.145	.038	.252	.075	.171	0.000	0.000	.066	.035	.076	-.028
0.000	0.000	.101	.147	.297	.000	.130	.041	.136	-.029	.136	-.116
-.329	.039	.185	.086	.400	-.036	.298	-.042	.214	-.063	.221	-.167
-.172	.059	.398	.026	.604	-.055	.397	-.089	.292	-.078	.295	-.154
-.030	-.041	.737	.154	.785	.175	.501	-.098	.403	-.053	.396	-.207
.128	-.103			.967	.161	.603	.015	.489	-.114	.497	-.265
.418	.013			1.000	-.118	.703	.034	.594	-.235	.597	-.189
.564	.062					.784	.113	.700	.001	.702	-.036
.710	.173					.868	.214	.786	.111	.786	.052
.976	.323					.923	.274	.858	.186	.864	.112
1.072	.272							.919	.202	.912	.114
1.110	.178							.967	.028		
0.000	0.000										
CN=	.6602		.7143		.7573		.8472		.6728		.5546
CM=	-.0284		-.0656		-.1044		-.1661		-.1106		-.0759

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Continued.

$\alpha = 6.95^{\circ}$; $C_L = 0.809$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.054	-.021	-1.032	.023	-1.014	.025	-.963	.022	-.888	.018	-.823
-.567	-.237	.035	-1.171	.068	-1.105	.079	-1.021	.075	-.994	.077	-.953
-.452	-.308	.105	-1.128	.134	-1.062	.133	-1.023	.129	-.981	.129	-.961
-.311	-.409	.178	-1.091	.209	-1.051	.214	-1.010	.201	-.983	.209	-.943
-.023	-.500	.286	-.821	.294	-1.022	.295	-.967	.294	-.934	.293	-.908
.133	-.404	.396	-.577	.404	-.988	.407	-.958	.397	-.921	.494	-.891
.272	-.345	.514	-.427	.497	-.902	.502	-.949	.495	-.901	.590	-.847
.416	-.288	.618	-.365	.599	-.624	.601	-.950	.594	-.849	.693	-.506
.565	-.178	.733	-.374	.700	-.550	.698	-.918	.693	-.487	.777	-.409
.713	-.156	.835	-.371	.864	-.328	.863	-.484	.784	-.417	.861	-.308
.854	-.226	.919	-.217	.926	-.243	.923	-.466	.856	-.358	.918	-.286
.980	-.282	.987	-.109	.975	-.214	.977	-.445	.926	-.328	.972	-.237
1.074	-.361							.977	-.314		
1.122	-.286										
					LOWER SURFACE						
-.660	.160	-.022	.456	.024	.454	.025	.378	.019	.324	.020	.314
-.616	.168	.038	.292	.075	.229	0.000	0.000	.066	.097	.076	.026
0.000	0.000	.101	.209	.297	.043	.130	.084	.136	.024	.136	-.073
-.329	.076	.185	.135	.400	-.007	.298	-.011	.214	-.033	.221	-.108
-.172	.091	.398	.064	.604	-.044	.397	-.063	.292	-.040	.295	-.140
-.030	-.002	.737	.169	.785	.173	.501	-.080	.403	-.034	.396	-.192
.128	-.052			.967	.141	.603	.015	.489	-.103	.497	-.258
.418	.046			1.000	-.194	.703	.024	.594	-.228	.597	-.201
.564	.093					.784	.105	.700	-.010	.702	-.055
.710	.199					.868	.210	.786	.109	.786	.031
.976	.337					.923	.268	.858	.187	.864	.088
1.072	.278							.919	.199	.912	.097
1.110	.179							.967	.018		
0.000	0.000										
CN=	.7425		.8150		.8495		.9225		.7466		.6494
CM=	-.0307		-.0726		-.1178		-.1791		-.1204		-.0997

$\alpha = 8.06^{\circ}$; $C_L = 0.899$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.107	-.021	-1.094	.023	-1.080	.025	-1.033	.022	-.959	.018	-.890
-.567	-.303	.035	-1.228	.068	-1.168	.079	-1.091	.075	-1.063	.077	-1.022
-.452	-.348	.105	-1.193	.134	-1.134	.133	-1.078	.129	-1.044	.129	-1.020
-.311	-.435	.178	-1.160	.209	-1.105	.214	-1.061	.201	-1.035	.209	-1.003
-.023	-.518	.286	-.951	.294	-1.065	.295	-1.015	.294	-1.002	.293	-.963
.133	-.415	.396	-.798	.404	-1.032	.407	-1.023	.397	-.988	.494	-.921
.272	-.359	.514	-.592	.497	-.947	.502	-1.013	.495	-.965	.590	-.860
.416	-.303	.618	-.449	.599	-.739	.601	-.966	.594	-.916	.693	-.520
.565	-.208	.733	-.401	.700	-.625	.698	-.668	.693	-.656	.777	-.439
.713	-.187	.835	-.409	.864	-.390	.863	-.555	.784	-.486	.861	-.419
.854	-.251	.919	-.286	.926	-.329	.923	-.527	.856	-.377	.918	-.406
.980	-.319	.987	-.147	.975	-.293	.977	-.497	.926	-.356	.972	-.401
1.074	-.396							.977	-.346		
1.122	-.362										
					LOWER SURFACE						
-.660	.172	-.022	.483	.024	.506	.025	.428	.019	.366	.020	.356
-.616	.201	.038	.359	.075	.294	0.000	0.000	.066	.138	.076	.079
0.000	0.000	.101	.245	.297	.084	.130	.126	.136	.054	.136	-.030
-.329	.110	.185	.190	.400	.034	.298	.024	.214	-.001	.221	-.082
-.172	.112	.398	.101	.604	-.033	.397	-.051	.292	-.025	.295	-.128
-.030	.034	.737	.181	.785	.167	.501	-.093	.403	-.027	.396	-.175
.128	-.012			.967	.105	.603	.008	.489	-.095	.497	-.235
.418	.089			1.000	-.318	.703	.008	.594	-.218	.597	-.195
.564	.127					.784	.084	.700	-.003	.702	-.063
.710	.226					.868	.215	.786	.106	.786	.015
.976	.357					.923	.275	.858	.184	.864	.069
1.072	.285							.919	.197	.912	.063
1.110	.184							.967	.014		
0.000	0.000										
CN=	.8431		.9483		.9370		.9490		.8272		.7149
CM=	-.0287		-.0927		-.1330		-.1718		-.1360		-.1124

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^0$

(f) M = 0.97. Continued.

$\alpha = 9.16^0$; $C_L = 0.981$

STA	.133	STA	.3C7	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.141	-.021	-1.157	.023	-1.141	.025	-1.096	.022	-1.027	.018	-.956
-.567	-.364	.035	-1.267	.068	-1.219	.079	-1.139	.075	-1.121	.077	-1.079
-.452	-.374	.105	-1.220	.134	-1.182	.133	-1.129	.129	-1.101	.129	-1.077
-.311	-.457	.178	-1.201	.209	-1.150	.214	-1.114	.201	-1.090	.209	-1.048
-.023	-.524	.286	-.992	.294	-1.116	.295	-1.080	.294	-1.056	.293	-1.009
.133	-.434	.396	-.887	.404	-1.056	.407	-1.079	.397	-1.030	.494	-.884
.272	-.367	.514	-.772	.497	-.856	.502	-1.052	.495	-.999	.590	-.615
.416	-.319	.618	-.733	.599	-.717	.601	-.996	.594	-.940	.693	-.559
.565	-.242	.733	-.491	.700	-.636	.698	-.864	.693	-.709	.777	-.538
.713	-.211	.835	-.465	.864	-.416	.863	-.612	.784	-.562	.861	-.470
.854	-.282	.919	-.334	.926	-.399	.923	-.546	.856	-.471	.918	-.437
.980	-.344	.987	-.190	.975	-.372	.977	-.441	.926	-.460	.972	-.432
1.074	-.425							.977	-.445		
1.122	-.442										
					LOWER SURFACE						
-.660	.184	-.022	.503	.024	.539	.025	.470	.019	.413	.020	.395
-.616	.218	.038	.398	.075	.341	0.000	0.000	.066	.186	.076	.119
0.000	0.000	.101	.308	.297	.119	.130	.173	.136	.089	.136	.019
-.329	.156	.185	.233	.400	.065	.298	.048	.214	.031	.221	-.057
-.172	.142	.398	.132	.604	-.021	.397	-.033	.292	-.004	.295	-.098
-.030	.080	.737	.197	.785	.154	.501	-.080	.403	-.016	.396	-.159
.128	.037			.967	.079	.603	.009	.489	-.083	.497	-.222
.418	.123			1.000	-.415	.703	-.004	.594	-.209	.597	-.199
.564	.156					.784	.070	.700	.004	.702	-.070
.710	.250					.868	.212	.786	.105	.786	.007
.976	.369					.923	.277	.858	.188	.864	.056
1.072	.292							.919	.193	.912	.051
1.110	.183							.967	-.001		
C.000	0.000										
CN=	.9375		1.0784		.9733		1.0260		.9006		.7356
CM=	-.0255		-.1240		-.1337		-.1857		-.1529		-.1089

$\alpha = 10.18^0$; $C_L = 1.043$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.192	-.021	-1.213	.023	-1.179	.025	-1.148	.022	-1.087	.018	-1.026
-.567	-.425	.035	-1.254	.068	-1.244	.079	-1.191	.075	-1.167	.077	-1.023
-.452	-.406	.105	-1.244	.134	-1.202	.133	-1.188	.129	-1.153	.129	-1.068
-.311	-.479	.178	-1.159	.209	-1.180	.214	-1.150	.201	-1.142	.209	-.924
-.023	-.538	.286	-1.026	.294	-1.103	.295	-1.111	.294	-1.099	.293	-.907
.133	-.435	.396	-1.019	.404	-.907	.407	-1.096	.397	-1.075	.494	-.718
.272	-.378	.514	-.960	.497	-.831	.502	-1.054	.495	-1.037	.590	-.654
.416	-.332	.618	-.912	.599	-.722	.601	-.857	.594	-.989	.693	-.603
.565	-.271	.733	-.836	.700	-.630	.698	-.816	.693	-.792	.777	-.569
.713	-.243	.835	-.554	.864	-.503	.863	-.478	.784	-.666	.861	-.528
.854	-.321	.919	-.332	.926	-.485	.923	-.431	.856	-.642	.918	-.490
.980	-.370	.987	-.176	.975	-.457	.977	-.371	.926	-.616	.972	-.466
1.074	-.441							.977	-.554		
1.122	-.514										
					LOWER SURFACE						
-.660	.196	-.022	.498	.024	.563	.025	.519	.019	.448	.020	.427
-.616	.261	.038	.445	.075	.390	0.000	0.000	.066	.228	.076	.166
0.000	0.000	.101	.358	.297	.159	.130	.213	.136	.133	.136	.054
-.329	.183	.185	.271	.400	.097	.298	.076	.214	.055	.221	-.023
-.172	.175	.398	.167	.604	-.003	.397	.002	.292	.024	.295	-.079
-.030	.116	.737	.209	.785	.160	.501	-.058	.403	-.004	.396	-.146
.128	.078			.967	.065	.603	.018	.489	-.074	.497	-.216
.418	.161			1.000	-.536	.703	-.003	.594	-.201	.597	-.194
.564	.186					.784	.047	.700	.009	.702	-.081
.710	.276					.868	.205	.786	.109	.786	-.001
.976	.383					.923	.267	.858	.184	.864	.046
1.072	.306							.919	.191	.912	.038
1.110	.198							.967	-.000		
0.000	0.000										
CN=	1.0314		1.2110		1.0022		1.0148		.9894		.7154
CM=	-.0235		-.1631		-.1423		-.1590		-.1800		-.1102

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97. Concluded.

$\alpha = 11.19^{\circ}$; $C_L = 1.093$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE							
	-.660	-.250	-.021	-1.244	.023	-1.141	.025	-1.201	.022	-1.132	.018	-1.081
	-.567	-.464	.035	-1.244	.068	-1.142	.079	-1.180	.075	-1.121	.077	-1.168
	-.452	-.437	.105	-1.243	.134	-1.102	.133	-1.163	.129	-1.082	.129	-1.127
	-.311	-.504	.178	-1.181	.209	-1.008	.214	-1.063	.201	-1.088	.209	-1.101
	-.023	-.538	.286	-1.126	.294	-.943	.295	-.934	.294	-.949	.293	-1.064
	.133	-.437	.396	-1.113	.404	-.873	.407	-.613	.397	-.893	.494	-.771
	.272	-.386	.514	-1.111	.497	-.813	.502	-.458	.495	-.855	.590	-.729
	.416	-.349	.618	-1.111	.599	-.745	.601	-.431	.594	-.819	.693	-.667
	.565	-.285	.733	-1.000	.700	-.697	.698	-.484	.693	-.747	.777	-.631
	.713	-.270	.835	-.633	.864	-.639	.863	-.475	.784	-.641	.861	-.585
	.854	-.348	.919	-.316	.926	-.617	.923	-.470	.856	-.590	.918	-.541
	.980	-.337	.987	-.122	.975	-.564	.977	-.467	.926	-.569	.972	-.558
	1.074	-.466							.977	-.533		
	1.122	-.522										
					LOWER SURFACE							
	-.660	.198	-.022	.507	.024	.605	.025	.545	.019	.469	.020	.454
	-.616	.277	.038	.474	.075	.442	0.000	0.000	.066	.276	.076	.201
	0.000	0.000	.101	.401	.297	.192	.130	.253	.136	.158	.136	.088
	-.329	.224	.185	.313	.400	.129	.298	.100	.214	.089	.221	-.001
	-.172	.218	.398	.209	.604	.023	.397	.028	.292	.047	.295	-.048
	-.030	.155	.737	.233	.785	.161	.501	-.026	.403	.012	.396	-.124
	.128	.119			.967	.063	.603	.035	.489	-.059	.497	-.202
	.418	.187			1.000	-.589	.703	.010	.594	-.187	.597	-.192
	.564	.219					.784	.049	.700	.007	.702	-.101
	.710	.301					.868	.202	.786	.110	.786	-.016
	.976	.405					.923	.264	.858	.183	.864	.036
	1.072	.318							.919	.193	.912	.022
	1.110	.204							.967	-.008		
	0.000	0.000										
CN=	1.1150		1.3302		1.0167		.8241		.9129		.8139	
CM=	-.0194		-.1921		-.1676		-.1076		-.1591		-.1227	

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^0$

(g) M = 0.98.

$\alpha = -1.11^0$; $C_L = -0.126$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	.018	-.021	.027	.023	-.001	.025	.048	.022	.191	.018	.296
-.567	-.003	.035	-.092	.068	-.147	.079	-.079	.075	.014	.077	.030
-.452	-.110	.105	-.129	.134	-.173	.133	-.109	.129	-.037	.129	-.033
-.311	-.217	.178	-.172	.209	-.158	.214	-.121	.201	-.056	.209	-.069
-.023	-.326	.286	-.164	.294	-.150	.295	-.092	.294	-.064	.293	-.112
.133	-.241	.396	-.170	.404	-.148	.407	-.109	.397	-.086	.494	-.211
.272	-.181	.514	-.124	.497	-.174	.502	-.101	.495	-.146	.590	-.262
.416	-.020	.618	-.161	.599	-.195	.601	-.159	.594	-.213	.693	-.351
.565	.010	.733	-.161	.700	-.261	.698	-.246	.693	-.298	.777	-.453
.713	.003	.835	-.201	.864	-.196	.863	-.453	.784	-.373	.861	-.200
.854	-.052	.919	-.091	.926	-.122	.923	-.164	.856	-.307	.918	-.036
.980	-.087	.987	.034	.975	-.029	.977	-.016	.926	-.205	.972	.032
1.074	-.102							.977	-.048		
1.122	-.013										
LOWER SURFACE											
-.660	.032	-.022	-.046	.024	-.399	.025	-.630	.019	-.669	.020	-.734
-.616	-.027	.038	-.323	.075	-.534	0.000	0.000	.066	-.877	.076	-.838
0.000	0.000	.101	-.382	.297	-.566	.130	-.787	.136	-.831	.136	-.816
-.329	-.125	.185	-.363	.400	-.569	.298	-.556	.214	-.807	.221	-.837
-.172	-.058	.398	-.439	.604	-.145	.397	-.262	.292	-.735	.295	-.702
-.030	-.193	.737	.063	.785	.157	.501	-.236	.403	-.298	.396	-.545
.128	-.315			.967	.223	.603	-.004	.489	-.199	.497	-.380
.418	-.280			1.000	.035	.703	.074	.594	-.169	.597	-.253
.564	-.244					.784	.140	.700	-.022	.702	-.105
.710	-.040					.868	.187	.786	.076	.786	.092
.976	.217					.923	.208	.858	.151	.864	.175
1.072	.226							.919	.188	.912	.207
1.110	.186							.967	.116		
0.000	0.000										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.0468	-.0797	-.0806	-.0817	-.1664	-.2079
CM=	-.0117	-.0415	-.0686	-.0960	-.0934	-.0812

$\alpha = -0.15^0$; $C_L = -0.003$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	.032	-.021	-.101	.023	-.120	.025	-.101	.022	.100	.018	.220
-.567	-.028	.035	-.320	.068	-.251	.079	-.190	.075	-.100	.077	-.055
-.452	-.143	.105	-.286	.134	-.250	.133	-.193	.129	-.097	.129	-.077
-.311	-.236	.178	-.344	.209	-.257	.214	-.188	.201	-.113	.209	-.114
-.023	-.347	.286	-.120	.294	-.228	.295	-.166	.294	-.104	.293	-.149
.133	-.263	.396	-.207	.404	-.175	.407	-.158	.397	-.109	.494	-.243
.272	-.220	.514	-.172	.497	-.205	.502	-.167	.495	-.168	.590	-.308
.416	-.133	.618	-.195	.599	-.232	.601	-.173	.594	-.230	.693	-.399
.565	-.016	.733	-.188	.700	-.289	.698	-.240	.693	-.318	.777	-.493
.713	-.009	.835	-.211	.864	-.209	.863	-.436	.784	-.401	.861	-.206
.854	-.068	.919	-.083	.926	-.122	.923	-.169	.856	-.496	.918	-.034
.980	-.111	.987	.029	.975	-.033	.977	-.019	.926	-.199	.972	.033
1.074	-.111							.977	-.057		
1.122	-.016										
LOWER SURFACE											
-.660	.065	-.022	.018	.024	-.264	.025	-.431	.019	-.520	.020	-.550
-.616	-.007	.038	-.246	.075	-.437	0.000	0.000	.066	-.748	.076	-.749
0.000	0.000	.101	-.320	.297	-.419	.130	-.542	.136	-.676	.136	-.707
-.329	-.099	.185	-.315	.400	-.086	.298	-.337	.214	-.635	.221	-.709
-.172	-.047	.398	-.394	.604	-.175	.397	-.312	.292	-.237	.295	-.439
-.030	-.169	.737	.111	.785	.169	.501	-.240	.403	-.248	.396	-.387
.128	-.302			.967	.224	.603	-.005	.489	-.164	.497	-.242
.418	-.261			1.000	.025	.703	.076	.594	-.236	.597	-.065
.564	-.212					.784	.118	.700	-.004	.702	.057
.710	.004					.868	.187	.786	.110	.786	.125
.976	.236					.923	.189	.858	.178	.864	.179
1.072	.250							.919	.216	.912	.190
1.110	.202							.967	.132		
0.000	0.000										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.1384	.0349	.0797	.0303	-.0240	-.0495
CM=	-.0169	-.0480	-.0783	-.0857	-.1039	-.1025

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(g) M = 0.98. *Continued.*

$\alpha = 0.90^o$; $C_L = 0.126$.

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.026	-.021	-.284	.023	-.356	.025	-.266	.022	-.099	.018	.060
-.567	-.023	.035	-.504	.068	-.492	.079	-.300	.075	-.278	.077	-.184
-.452	-.158	.105	-.346	.134	-.419	.133	-.280	.129	-.259	.129	-.178
-.311	-.256	.178	-.392	.209	-.287	.214	-.247	.201	-.240	.209	-.191
-.023	-.370	.286	-.367	.294	-.170	.295	-.244	.294	-.205	.293	-.190
.133	-.299	.396	-.270	.404	-.177	.407	-.235	.397	-.176	.494	-.261
.272	-.252	.514	-.180	.497	-.227	.502	-.270	.495	-.219	.590	-.315
.416	-.204	.618	-.189	.599	-.278	.601	-.307	.594	-.262	.693	-.404
.565	-.060	.733	-.200	.700	-.314	.698	-.325	.693	-.321	.777	-.482
.713	-.044	.835	-.198	.864	-.240	.863	-.425	.784	-.401	.861	-.148
.854	-.076	.919	-.093	.926	-.111	.923	-.149	.856	-.500	.918	-.004
.980	-.119	.987	.025	.975	-.035	.977	-.014	.926	-.189	.972	.040
1.074	-.111							.977	-.065		
1.122	-.023										
				LOWER SURFACE							
-.660	.087	-.022	.085	.024	-.096	.025	-.217	.019	-.293	.020	-.300
-.616	.020	.038	-.187	.075	-.353	0.000	0.000	.066	-.531	.076	-.551
0.000	0.000	.101	-.248	.297	-.236	.130	-.266	.136	-.460	.136	-.525
-.329	-.082	.185	-.269	.400	-.230	.298	-.188	.214	-.380	.221	-.372
-.172	-.030	.398	-.139	.604	-.157	.397	-.270	.292	-.317	.295	-.256
-.030	-.153	.737	.112	.785	.165	.501	-.256	.403	-.251	.396	-.375
.128	-.278			.967	.216	.603	-.003	.489	-.125	.497	-.266
.418	-.246			1.000	.005	.703	.072	.594	-.224	.597	-.036
.564	-.130					.784	.126	.700	.031	.702	.091
.710	.049					.868	.186	.786	.151	.786	.147
.976	.265					.923	.199	.858	.220	.864	.182
1.072	.265							.919	.235	.912	.192
1.110	.202							.967	.140		
0.000	0.000										
CN=	.2291		.1921		.1564		.1805		.1074		.0676
CM=	-.0267		-.0527		-.0719		-.0875		-.1021		-.0924

$\alpha = 1.94^o$; $C_L = 0.252$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.018	-.021	-.437	.023	-.515	.025	-.481	.022	-.284	.018	-.159
-.567	-.049	.035	-.684	.068	-.672	.079	-.586	.075	-.421	.077	-.370
-.452	-.180	.105	-.454	.134	-.553	.133	-.552	.129	-.380	.129	-.363
-.311	-.286	.178	-.465	.209	-.509	.214	-.501	.201	-.327	.209	-.305
-.023	-.403	.286	-.396	.294	-.453	.295	-.166	.294	-.284	.293	-.289
.133	-.313	.396	-.374	.404	-.220	.407	-.264	.397	-.266	.494	-.294
.272	-.276	.514	-.248	.497	-.228	.502	-.298	.495	-.310	.590	-.339
.416	-.234	.618	-.236	.599	-.310	.601	-.335	.594	-.362	.693	-.410
.565	-.121	.733	-.227	.700	-.336	.698	-.410	.693	-.408	.777	-.499
.713	-.064	.835	-.245	.864	-.232	.863	-.263	.784	-.440	.861	-.099
.854	-.107	.919	-.118	.926	-.111	.923	-.118	.856	-.242	.918	-.001
.980	-.150	.987	.002	.975	-.046	.977	-.037	.926	-.154	.972	.049
1.074	-.159							.977	-.057		
1.122	-.055										
				LOWER SURFACE							
-.660	.100	-.022	.135	.024	.051	.025	-.083	.019	-.132	.020	-.074
-.616	.052	.038	-.105	.075	-.137	0.000	0.000	.066	-.324	.076	-.260
0.000	0.000	.101	-.145	.297	-.183	.130	-.203	.136	-.205	.136	-.326
-.329	-.059	.185	-.201	.400	-.166	.298	-.215	.214	-.189	.221	-.317
-.172	-.025	.398	-.117	.604	-.124	.397	-.264	.292	-.186	.295	-.322
-.030	-.145	.737	.118	.785	.169	.501	-.219	.403	-.240	.396	-.241
.128	-.262			.967	.212	.603	.028	.489	-.122	.497	-.260
.418	-.200			1.000	-.008	.703	.074	.594	-.203	.597	-.064
.564	-.053					.784	.148	.700	.047	.702	.088
.710	.086					.868	.215	.786	.156	.786	.170
.976	.274					.923	.248	.858	.224	.864	.208
1.072	.258							.919	.244	.912	.216
1.110	.191							.967	.128		
0.000	0.000										
CN=	.3237		.3007		.2996		.2689		.2355		.1870
CM=	-.0347		-.0571		-.0684		-.0775		-.0931		-.0847

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(g) M = 0.98. Continued.

$\alpha = 2.41^{\circ}$; $C_L = 0.312$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.028	-.021	-.533	.023	-.556	.025	-.544	.022	-.386	.018	-.311
-.567	-.066	.035	-.719	.068	-.727	.079	-.647	.075	-.599	.077	-.546
-.452	-.198	.105	-.550	.134	-.653	.133	-.655	.129	-.588	.129	-.422
-.311	-.315	.178	-.488	.209	-.619	.214	-.628	.201	-.552	.209	-.368
-.023	-.417	.286	-.436	.294	-.537	.295	-.523	.294	-.531	.293	-.344
.133	-.331	.396	-.390	.404	-.274	.407	-.208	.397	-.200	.494	-.347
.272	-.286	.514	-.267	.497	-.233	.502	-.295	.495	-.296	.590	-.353
.416	-.252	.618	-.271	.599	-.300	.601	-.335	.594	-.357	.693	-.418
.565	-.136	.733	-.267	.700	-.358	.698	-.417	.693	-.438	.777	-.194
.713	-.073	.835	-.251	.864	-.264	.863	-.367	.784	-.524	.861	-.022
.854	-.121	.919	-.118	.926	-.123	.923	-.132	.856	-.364	.918	-.042
.980	-.168	.987	-.011	.975	-.064	.977	-.059	.926	-.116	.972	.059
1.074	-.183							.977	-.051		
1.122	-.078										
				LOWER SURFACE							
-.660	.113	-.022	.211	.024	.117	.025	.022	.019	-.069	.020	.024
-.616	.077	.038	-.025	.075	-.085	0.000	0.000	.066	-.277	.076	-.230
0.000	0.000	.101	-.065	.297	-.148	.130	-.138	.136	-.194	.136	-.287
-.329	-.045	.185	-.119	.400	-.136	.298	-.212	.214	-.184	.221	-.277
-.172	-.012	.398	-.100	.604	-.083	.397	-.259	.292	-.202	.295	-.295
-.030	-.127	.737	.130	.785	.170	.501	-.199	.403	-.242	.396	-.220
.128	-.249			.967	.205	.603	.021	.489	-.093	.497	-.210
.418	-.159			1.000	-.025	.703	.056	.594	-.196	.597	-.019
.564	-.029					.784	.133	.700	.056	.702	.123
.710	.100					.868	.212	.786	.167	.786	.203
.976	.285					.923	.255	.858	.238	.864	.240
1.072	.265							.919	.263	.912	.242
1.110	.193							.967	.153		
0.000	0.000										
CN=	.3828		.3752		.3694		.3520		.3268		.2389
CM=	-.0354		-.0590		-.0733		-.0806		-.0985		-.0755

$\alpha = 2.92^{\circ}$; $C_L = 0.378$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.022	-.021	-.631	.023	-.643	.025	-.606	.022	-.457	.018	-.387
-.567	-.080	.035	-.747	.068	-.793	.079	-.691	.075	-.669	.077	-.612
-.452	-.205	.105	-.603	.134	-.754	.133	-.710	.129	-.659	.129	-.646
-.311	-.315	.178	-.521	.209	-.699	.214	-.716	.201	-.668	.209	-.607
-.023	-.418	.286	-.450	.294	-.637	.295	-.645	.294	-.564	.293	-.578
.133	-.344	.396	-.402	.404	-.458	.407	-.360	.397	-.599	.494	-.485
.272	-.297	.514	-.284	.497	-.245	.502	-.281	.495	-.255	.590	-.331
.416	-.263	.618	-.294	.599	-.286	.601	-.324	.594	-.330	.693	-.409
.565	-.128	.733	-.280	.700	-.355	.698	-.415	.693	-.429	.777	-.054
.713	-.080	.835	-.269	.864	-.259	.863	-.437	.784	-.520	.861	-.039
.854	-.137	.919	-.123	.926	-.134	.923	-.143	.856	-.393	.918	-.042
.980	-.173	.987	-.013	.975	-.067	.977	-.083	.926	-.112	.972	.060
1.074	-.205							.977	-.065		
1.122	-.101										
				LOWER SURFACE							
-.660	.120	-.022	.258	.024	.170	.025	.101	.019	.015	.020	.083
-.616	.080	.038	.026	.075	-.044	0.000	0.000	.066	-.228	.076	-.192
0.000	0.000	.101	-.031	.297	-.123	.130	-.122	.136	-.162	.136	-.252
-.329	-.036	.185	-.083	.400	-.122	.298	-.201	.214	-.168	.221	-.266
-.172	-.004	.398	-.085	.604	-.054	.397	-.207	.292	-.189	.295	-.292
-.030	-.119	.737	.131	.785	.168	.501	-.193	.403	-.236	.396	-.217
.128	-.244			.967	.203	.603	.009	.489	-.099	.497	-.223
.418	-.091			1.000	-.037	.703	.047	.594	-.212	.597	-.063
.564	-.019					.784	.118	.700	.039	.702	.101
.710	.110					.868	.207	.786	.160	.786	.186
.976	.286					.923	.251	.858	.226	.864	.232
1.072	.262							.919	.260	.912	.240
1.110	.185							.967	.139		
0.000	0.000										
CN=	.4182		.4188		.4382		.4145		.3916		.3197
CM=	-.0386		-.0589		-.0744		-.0846		-.0968		-.0659

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(g) M = 0.98. Continued.

$\alpha = 3.41^o$; $C_L = 0.438$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	.015	-.021	-.701	.023	-.699	.025	-.644	.022	-.522	.018	-.445
-.567	-.093	.035	-.757	.068	-.821	.079	-.741	.075	-.698	.077	-.667
-.452	-.220	.105	-.715	.134	-.793	.133	-.753	.129	-.705	.129	-.679
-.311	-.323	.178	-.537	.209	-.779	.214	-.753	.201	-.709	.209	-.661
-.023	-.436	.286	-.474	.294	-.703	.295	-.700	.294	-.659	.293	-.635
.133	-.351	.396	-.438	.404	-.535	.407	-.680	.397	-.657	.494	-.608
.272	-.314	.514	-.314	.497	-.263	.502	-.335	.495	-.623	.590	-.603
.416	-.279	.618	-.306	.599	-.282	.601	-.323	.594	-.338	.693	-.399
.565	-.128	.733	-.289	.700	-.372	.698	-.411	.693	-.418	.777	-.058
.713	-.089	.835	-.281	.864	-.310	.863	-.371	.784	-.505	.861	-.009
.854	-.142	.919	-.131	.926	-.144	.923	-.147	.856	-.340	.918	-.005
.980	-.183	.987	-.029	.975	-.075	.977	-.103	.926	-.129	.972	.067
1.074	-.226							.977	-.088		
1.122	-.117										
					LOWER SURFACE						
-.660	.129	-.022	.308	.024	.228	.025	.121	.019	.065	.020	.120
-.616	.090	.038	.077	.075	.003	0.000	0.000	.066	-.151	.076	-.160
0.000	0.000	.101	-.003	.297	-.114	.130	-.092	.136	-.139	.136	-.234
-.329	-.020	.185	-.038	.400	-.093	.298	-.183	.214	-.146	.221	-.244
-.172	.006	.398	-.065	.604	-.054	.397	-.102	.292	-.175	.295	-.275
-.030	-.106	.737	.133	.785	.171	.501	-.189	.403	-.201	.396	-.195
.128	-.225			.967	.199	.603	.018	.489	-.091	.497	-.217
.418	-.071			1.000	-.050	.703	.048	.594	-.227	.597	-.083
.564	-.006					.784	.118	.700	.030	.702	.073
.710	.123					.868	.213	.786	.154	.786	.167
.976	.295					.923	.264	.858	.227	.864	.208
1.072	.254							.919	.251	.912	.226
1.110	.187							.967	.127		
0.000	0.000										

	.133	.307	.480	.653	.804	.933
CN=	.4606	.4711	.4910	.4834	.4625	.3820
CM=	-.0379	-.0599	-.0789	-.0882	-.1011	-.0700

$\alpha = 3.93^o$; $C_L = 0.500$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	.011	-.021	-.750	.023	-.741	.025	-.710	.022	-.569	.018	-.501
-.567	-.116	.035	-.880	.068	-.872	.079	-.783	.075	-.739	.077	-.707
-.452	-.234	.105	-.838	.134	-.843	.133	-.800	.129	-.736	.129	-.730
-.311	-.331	.178	-.551	.209	-.824	.214	-.784	.201	-.757	.209	-.703
-.023	-.438	.286	-.481	.294	-.765	.295	-.741	.294	-.701	.293	-.670
.133	-.360	.396	-.463	.404	-.711	.407	-.725	.397	-.695	.494	-.656
.272	-.312	.514	-.326	.497	-.328	.502	-.714	.495	-.655	.590	-.638
.416	-.266	.618	-.314	.599	-.290	.601	-.365	.594	-.652	.693	-.354
.565	-.136	.733	-.298	.700	-.397	.698	-.410	.693	-.636	.777	-.136
.713	-.094	.835	-.292	.864	-.327	.863	-.340	.784	-.492	.861	-.073
.854	-.150	.919	-.130	.926	-.146	.923	-.151	.856	-.259	.918	-.056
.980	-.201	.987	-.037	.975	-.083	.977	-.111	.926	-.148	.972	-.052
1.074	-.247							.977	-.120		
1.122	-.132										
					LOWER SURFACE						
-.660	.142	-.022	.338	.024	.277	.025	.183	.019	.097	.020	.156
-.616	.108	.038	.123	.075	.031	0.000	0.000	.066	-.120	.076	-.132
0.000	0.000	.101	.032	.297	-.088	.130	-.074	.136	-.122	.136	-.212
-.329	-.008	.185	-.009	.400	-.088	.298	-.164	.214	-.137	.221	-.236
-.172	.014	.398	-.043	.604	-.057	.397	-.103	.292	-.159	.295	-.268
-.030	-.099	.737	.136	.785	.175	.501	-.196	.403	-.127	.396	-.190
.128	-.221			.967	.195	.603	.020	.489	-.105	.497	-.242
.418	-.056			1.000	-.055	.703	.043	.594	-.233	.597	-.138
.564	.005					.784	.116	.700	.029	.702	.023
.710	.132					.868	.211	.786	.147	.786	.110
.976	.299					.923	.267	.858	.222	.864	.157
1.072	.266							.919	.241	.912	.158
1.110	.186							.967	.105		
0.000	0.000										

	.133	.307	.480	.653	.804	.933
CN=	.4871	.5187	.5505	.5495	.5407	.4042
CM=	-.0371	-.0587	-.0843	-.0947	-.1169	-.0677

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(g) M = 0.98. Concluded.

$\alpha = 4.95^o$; $C_L = 0.614$

	STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
UPPER SURFACE												
	-.660	-.007	-.021	-.852	.023	-.836	.025	-.797	.022	-.681	.018	-.616
	-.567	-.165	.035	-.993	.068	-.952	.079	-.873	.075	-.833	.077	-.787
	-.452	-.251	.105	-.929	.134	-.922	.133	-.877	.129	-.815	.129	-.808
	-.311	-.347	.178	-.894	.209	-.910	.214	-.870	.201	-.830	.209	-.778
	-.023	-.454	.286	-.504	.294	-.877	.295	-.831	.294	-.785	.293	-.757
	.133	-.377	.396	-.517	.404	-.844	.407	-.805	.397	-.776	.494	-.747
	.272	-.328	.514	-.375	.497	-.623	.502	-.794	.495	-.759	.590	-.726
	.416	-.282	.618	-.344	.599	-.358	.601	-.780	.594	-.740	.693	-.726
	.565	-.159	.733	-.331	.700	-.438	.698	-.697	.693	-.763	.777	-.290
	.713	-.119	.835	-.318	.864	-.312	.863	-.271	.784	-.339	.861	-.213
	.854	-.180	.919	-.149	.926	-.153	.923	-.233	.856	-.319	.918	-.170
	.980	-.225	.987	-.061	.975	-.101	.977	-.178	.926	-.312	.972	-.149
	1.074	-.283							.977	-.306		
	1.122	-.185										
LOWER SURFACE												
	-.660	.150	-.022	.387	.024	.349	.025	.253	.019	.197	.020	.222
	-.616	.131	.038	.191	.075	.113	0.000	0.000	.066	-.019	.076	-.070
	0.000	0.000	.101	.104	.297	-.039	.130	-.005	.136	-.074	.136	-.170
	-.329	.022	.185	.047	.400	-.064	.298	-.052	.214	-.102	.221	-.196
	-.172	.040	.398	-.012	.604	-.053	.397	-.097	.292	-.129	.295	-.241
	-.030	-.071	.737	.153	.785	.181	.501	-.177	.403	-.031	.396	-.188
	.128	-.164			.967	.188	.603	.021	.489	-.099	.497	-.268
	.418	-.017			1.000	-.071	.703	.038	.594	-.233	.597	-.205
	.564	.037					.784	.112	.700	-.001	.702	-.043
	.710	.150					.868	.218	.786	.116	.786	.043
	.976	.318					.923	.274	.858	.186	.864	.102
	1.072	.272							.919	.202	.912	.101
	1.110	.187							.967	.034		
	0.000	0.000										
CN=	.5769		.6331		.6640		.7071		.6356		.5136	
CM=	-.0391		-.0632		-.0945		-.1269		-.1262		-.0910	

[redacted]

TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(h) M = 0.99.

$\alpha = -4.99^o$; $C_L = -0.551$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.055	-.021	.377	.023	.371	.025	.343	.022	.418	.018	.465
-.567	-.041	.035	.231	.068	.240	.079	.218	.075	.225	.077	.228
-.452	-.100	.105	.148	.134	.137	.133	.153	.129	.171	.129	.164
-.311	-.149	.178	.079	.209	.077	.214	.105	.201	.113	.209	.091
-.023	-.030	.286	.046	.294	.059	.295	.075	.294	.071	.293	.028
.133	.025	.396	.025	.404	.021	.407	.024	.397	.028	.494	-.097
.272	.056	.514	.030	.497	-.016	.502	-.024	.495	-.046	.590	-.176
.416	.087	.618	-.013	.599	-.046	.601	-.079	.594	-.104	.693	-.279
.565	.113	.733	-.048	.700	-.117	.698	-.166	.693	-.171	.777	-.380
.713	.107	.835	-.108	.864	-.223	.863	-.393	.784	-.249	.861	-.484
.854	.051	.919	-.087	.926	-.202	.923	-.231	.856	-.350	.918	-.630
.980	-.005	.987	.024	.975	-.061	.977	-.086	.926	-.524	.972	-.421
1.074	-.068							.977	-.627		
1.122	-.043										
				LOWER SURFACE							
-.660	-.161	-.022	-.564	.024	-.869	.025	-.943	.019	-1.008	.020	-.706
-.616	-.201	.038	-.814	.075	-.898	0.000	0.000	.066	-1.088	.076	-.639
C.000	0.000	.101	-.698	.297	-.901	.130	-1.045	.136	-1.032	.136	-.592
-.329	-.241	.185	-.588	.400	-.872	.298	-.994	.214	-1.035	.221	-.547
-.172	-.070	.398	-.539	.604	-.202	.397	-.943	.292	-.919	.295	-.516
-.030	-.195	.737	-.166	.785	.038	.501	-.542	.403	-.767	.396	-.466
.128	-.300			.967	.029	.603	-.252	.489	-.710	.497	-.454
.418	-.253			1.000	-.012	.703	-.093	.594	-.657	.597	-.422
.564	-.326					.784	.053	.700	-.576	.702	-.445
.710	-.248					.868	.056	.786	-.386	.786	-.437
.976	.164					.923	-.002	.858	-.262	.864	-.372
1.072	.207							.919	-.222	.912	-.375
1.110	.173							.967	-.222		
0.000	0.000										
CN=	-.2590		-.5086		-.4667		-.4906		-.6151		-.3563
CM=	.0023		.0035		-.0365		-.0498		.0020		-.0126

$\alpha = -3.96^o$; $C_L = -0.459$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.044	-.021	.330	.023	.291	.025	.273	.022	.365	.018	.430
-.567	-.048	.035	.158	.068	.151	.079	.129	.075	.178	.077	.189
-.452	-.111	.105	.078	.134	.077	.133	.099	.129	.117	.129	.121
-.311	-.172	.178	.028	.209	.040	.214	.044	.201	.062	.209	.063
-.023	-.200	.286	-.004	.294	.009	.295	.035	.294	.032	.293	.006
.133	.002	.396	-.023	.404	-.022	.407	-.010	.397	.000	.494	-.113
.272	.032	.514	-.013	.497	-.060	.502	-.055	.495	-.069	.590	-.189
.416	.057	.618	-.048	.599	-.091	.601	-.093	.594	-.134	.693	-.290
.565	.088	.733	-.084	.700	-.146	.698	-.176	.693	-.183	.777	-.389
.713	.073	.835	-.145	.864	-.194	.863	-.402	.784	-.213	.861	-.486
.854	.025	.919	-.096	.926	-.150	.923	-.152	.856	-.342	.918	-.637
.980	-.025	.987	.016	.975	-.030	.977	-.034	.926	-.442	.972	-.433
1.074	-.088							.977	-.457		
1.122	-.040										
				LOWER SURFACE							
-.660	-.100	-.022	-.438	.024	-.789	.025	-.867	.019	-.929	.020	-.688
-.616	-.137	.038	-.694	.075	-.833	0.000	0.000	.066	-1.054	.076	-.644
0.000	0.000	.101	-.587	.297	-.826	.130	-.992	.136	-.997	.136	-.615
-.329	-.223	.185	-.514	.400	-.851	.298	-.959	.214	-.981	.221	-.541
-.172	-.063	.398	-.516	.604	-.197	.397	-.921	.292	-.934	.295	-.531
-.030	-.205	.737	-.097	.785	.017	.501	-.627	.403	-.709	.396	-.442
.128	-.334			.967	.084	.603	-.170	.489	-.653	.497	-.409
.418	-.238			1.000	.021	.703	.083	.594	-.638	.597	-.377
.564	-.293					.784	.197	.700	-.529	.702	-.381
.710	-.176					.868	.189	.786	-.286	.786	-.376
.976	.174					.923	.084	.858	-.107	.864	-.340
1.072	.218							.919	-.085	.912	-.331
1.110	.180							.967	-.041		
0.000	0.000										
CN=	-.1620		-.3911		-.4051		-.4054		-.5415		-.3100
CM=	.0019		-.0125		-.0359		-.0673		-.0139		-.0271

[redacted]

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = -2.98^{\circ}$; $C_L = -0.358$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	-.043	-.021	.250	.023	.223	.025	.221	.022	.325	.018	.404
	-.567	-.054	.035	.070	.068	.061	.079	.074	.075	.122	.077	.144
	-.452	-.123	.105	.028	.134	.006	.133	.037	.129	.076	.129	.085
	-.311	-.188	.178	-.034	.209	-.031	.214	-.004	.201	.032	.209	.026
	-.023	-.263	.286	-.059	.294	-.054	.295	-.013	.294	.009	.293	-.023
	.133	-.089	.396	-.075	.404	-.067	.407	-.029	.397	-.022	.494	-.124
	.272	.008	.514	-.054	.497	-.107	.502	-.069	.495	-.089	.590	-.185
	.416	.024	.618	-.088	.599	-.132	.601	-.124	.594	-.149	.693	-.292
	.565	.061	.733	-.109	.700	-.167	.698	-.201	.693	-.219	.777	-.386
	.713	.058	.835	-.170	.864	-.199	.863	-.385	.784	-.259	.861	-.478
	.854	.001	.919	-.104	.926	-.128	.923	-.155	.856	-.329	.918	-.604
	.980	-.050	.987	.022	.975	.000	.977	-.004	.926	-.350	.972	-.386
	1.074	-.100							.977	-.189		
	1.122	-.039										
						LOWER SURFACE						
	-.660	-.059	-.022	-.303	.024	-.683	.025	-.799	.019	-.854	.020	-.621
	-.616	-.116	.038	-.532	.075	-.711	0.000	0.000	.066	-.987	.076	-.619
	0.000	0.000	.101	-.488	.297	-.731	.130	-.935	.136	-.930	.136	-.526
	-.329	-.181	.185	-.499	.400	-.722	.298	-.889	.214	-.911	.221	-.519
	-.172	-.065	.398	-.477	.604	-.190	.397	-.861	.292	-.843	.295	-.461
	-.030	-.197	.737	-.038	.785	.071	.501	-.613	.403	-.626	.396	-.394
	.128	-.335			.967	.112	.603	-.062	.489	-.540	.497	-.352
	.418	-.232			1.000	.059	.703	.117	.594	-.500	.597	-.317
	.564	-.270					.784	.229	.700	-.419	.702	-.316
	.710	-.141					.868	.226	.786	-.281	.786	-.312
	.976	.181					.923	.165	.858	-.050	.864	-.293
	1.072	.218							.919	.023	.912	-.272
	1.110	.179							.967	.074		
	0.000	0.000										
CN=		-.0856		-.2836		-.2950		-.3275		-.4490		-.2438
CM=		.0036		-.0248		-.0446		-.0774		-.0267		-.0374

$\alpha = -1.99^{\circ}$; $C_L = -0.250$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	-.029	-.021	.177	.023	.106	.025	.113	.022	.275	.018	.349
	-.567	-.059	.035	-.018	.068	-.015	.079	-.005	.075	.073	.077	.086
	-.452	-.135	.105	-.058	.134	-.066	.133	-.048	.129	.040	.129	.040
	-.311	-.206	.178	-.101	.209	-.074	.214	-.059	.201	-.002	.209	-.007
	-.023	-.279	.286	-.099	.294	-.101	.295	-.072	.294	-.017	.293	-.054
	.133	-.180	.396	-.103	.404	-.102	.407	-.075	.397	-.044	.494	-.164
	.272	-.029	.514	-.095	.497	-.137	.502	-.046	.495	-.110	.590	-.226
	.416	.006	.618	-.127	.599	-.170	.601	-.115	.594	-.173	.693	-.305
	.565	.038	.733	-.139	.700	-.246	.698	-.202	.693	-.253	.777	-.375
	.713	.035	.835	-.195	.864	-.150	.863	-.419	.784	-.340	.861	-.326
	.854	-.020	.919	-.117	.926	-.089	.923	-.143	.856	-.294	.918	-.473
	.980	-.065	.987	.026	.975	.009	.977	.005	.926	-.152	.972	-.319
	1.074	-.115							.977	.005		
	1.122	-.037										
						LOWER SURFACE						
	-.660	-.006	-.022	-.103	.024	-.550	.025	-.680	.019	-.756	.020	-.706
	-.616	-.095	.038	-.408	.075	-.605	0.000	0.000	.066	-.929	.076	-.708
	0.000	0.000	.101	-.448	.297	-.656	.130	-.855	.136	-.890	.136	-.727
	-.329	-.153	.185	-.443	.400	-.552	.298	-.813	.214	-.860	.221	-.556
	-.172	-.050	.398	-.439	.604	-.134	.397	-.778	.292	-.832	.295	-.458
	-.030	-.186	.737	.011	.785	.112	.501	-.139	.403	-.522	.396	-.429
	.128	-.316			.967	.174	.603	.034	.489	-.422	.497	-.326
	.418	-.242			1.000	.075	.703	.112	.594	-.329	.597	-.284
	.564	-.247					.784	.174	.700	-.122	.702	-.246
	.710	-.089					.868	.255	.786	.016	.786	-.233
	.976	.195					.923	.218	.858	.116	.864	-.216
	1.072	.213							.919	.165	.912	-.183
	1.110	.180							.967	.163		
	0.000	0.000										
CN=		-.0140		-.1767		-.1748		-.2001		-.3129		-.2355
CM=		.0014		-.0348		-.0579		-.0909		-.0648		-.0457

[redacted]

TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(h) M = 0.99. Continued.

$\alpha = -1.05^o$; $C_L = -0.134$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.021	-.021	.021	.023	-.008	.025	.029	.022	.200	.018	.299
-.567	-.067	.035	-.087	.068	-.147	.079	-.102	.075	.017	.077	.032
-.452	-.144	.105	-.137	.134	-.149	.133	-.110	.129	-.017	.129	-.006
-.311	-.218	.178	-.172	.209	-.175	.214	-.113	.201	-.032	.209	-.047
-.023	-.300	.286	-.149	.294	-.147	.295	-.123	.294	-.037	.293	-.089
.133	-.212	.396	-.168	.404	-.132	.407	-.130	.397	-.062	.494	-.202
.272	-.119	.514	-.130	.497	-.163	.502	-.161	.495	-.128	.590	-.262
.416	-.022	.618	-.159	.599	-.200	.601	-.115	.594	-.187	.693	-.352
.565	.011	.733	-.164	.700	-.272	.698	-.155	.693	-.275	.777	-.454
.713	.012	.835	-.204	.864	-.155	.863	-.412	.784	-.357	.861	-.277
.854	-.042	.919	-.118	.926	-.068	.923	-.194	.856	-.481	.918	-.081
.980	-.090	.987	.027	.975	.004	.977	-.012	.926	-.174	.972	.002
1.074	-.127							.977	-.036		
1.122	-.033										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.013	-.022	-.016	.024	-.386	.025	-.591	.019	-.640	.020	-.720
-.616	-.047	.038	-.311	.075	-.508	0.000	0.000	.066	-.837	.076	-.799
0.000	0.000	.101	-.378	.297	-.548	.130	-.769	.136	-.807	.136	-.756
-.329	-.120	.185	-.356	.400	-.556	.298	-.714	.214	-.775	.221	-.757
-.172	-.046	.398	-.415	.604	-.113	.397	-.557	.292	-.743	.295	-.598
-.030	-.177	.737	.040	.785	.158	.501	-.106	.403	-.359	.396	-.495
.128	-.305			.967	.233	.603	.042	.489	-.220	.497	-.351
.418	-.254			1.000	.049	.703	.116	.594	-.150	.597	-.260
.564	-.237					.784	.162	.700	-.015	.702	-.112
.710	-.052					.868	.228	.786	.080	.786	.026
.976	.205					.923	.237	.858	.163	.864	.093
1.072	.224							.919	.160	.912	.168
1.110	.180							.967	.129		
0.000	0.000										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.0513	-.0771	-.0743	-.1065	-.1710	-.1920
CM=	-.0016	-.0388	-.0658	-.0928	-.0948	-.0785

$\alpha = -0.05^o$; $C_L = -0.004$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.006	-.021	-.061	.023	-.205	.025	-.081	.022	.097	.018	.225
-.567	-.079	.035	-.293	.068	-.248	.079	-.175	.075	-.097	.077	-.037
-.452	-.157	.105	-.258	.134	-.240	.133	-.186	.129	-.106	.129	-.056
-.311	-.245	.178	-.307	.209	-.264	.214	-.191	.201	-.130	.209	-.096
-.023	-.332	.286	-.216	.294	-.237	.295	-.170	.294	-.098	.293	-.128
.133	-.247	.396	-.201	.404	-.188	.407	-.175	.397	-.074	.494	-.221
.272	-.183	.514	-.183	.497	-.204	.502	-.224	.495	-.125	.590	-.286
.416	-.113	.618	-.199	.599	-.249	.601	-.263	.594	-.196	.693	-.373
.565	-.028	.733	-.197	.700	-.288	.698	-.156	.693	-.282	.777	-.471
.713	-.018	.835	-.240	.864	-.163	.863	-.322	.784	-.367	.861	-.333
.854	-.071	.919	-.111	.926	-.086	.923	-.166	.856	-.494	.918	-.076
.980	-.118	.987	.032	.975	-.012	.977	.001	.926	-.186	.972	-.011
1.074	-.142							.977	-.050		
1.122	-.042										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.040	-.022	.040	.024	-.261	.025	-.430	.019	-.526	.020	-.539
-.616	-.029	.038	-.233	.075	-.427	0.000	0.000	.066	-.731	.076	-.691
0.000	0.000	.101	-.292	.297	-.473	.130	-.674	.136	-.662	.136	-.706
-.329	-.097	.185	-.306	.400	-.447	.298	-.164	.214	-.563	.221	-.742
-.172	-.036	.398	-.383	.604	-.123	.397	-.223	.292	-.301	.295	-.478
-.030	-.161	.737	.079	.785	.176	.501	-.244	.403	-.247	.396	-.339
.128	-.283			.967	.232	.603	.008	.489	-.176	.497	-.242
.418	-.238			1.000	.022	.703	.096	.594	-.192	.597	-.059
.564	-.211					.784	.147	.700	.019	.702	.056
.710	-.020					.868	.201	.786	.109	.786	.126
.976	.220					.923	.217	.858	.168	.864	.173
1.072	.230							.919	.201	.912	.175
1.110	.183							.967	.119		
0.000	0.000										

	STA .133	STA .307	STA .480	STA .653	STA .804	STA .933
CN=	.1439	.0379	.0291	.0414	-.0362	-.0512
CM=	-.0082	-.0463	-.0692	-.0860	-.0973	-.1068

[redacted]

TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = 0.94^{\circ}$; $C_L = 0.124$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.001	-.021	-.283	.023	-.322	.025	-.286	.022	-.078	.018	.037
-.567	-.092	.035	-.485	.068	-.473	.079	-.335	.075	-.256	.077	-.179
-.452	-.175	.105	-.328	.134	-.413	.133	-.340	.129	-.263	.129	-.168
-.311	-.252	.178	-.357	.209	-.405	.214	-.218	.201	-.263	.209	-.171
-.023	-.352	.286	-.348	.294	-.296	.295	-.210	.294	-.223	.293	-.190
.133	-.252	.396	-.295	.404	-.205	.407	-.227	.397	-.198	.494	-.257
.272	-.208	.514	-.217	.497	-.220	.502	-.257	.495	-.183	.590	-.305
.416	-.175	.618	-.234	.599	-.299	.601	-.293	.594	-.237	.693	-.354
.565	-.073	.733	-.244	.700	-.315	.698	-.370	.693	-.303	.777	-.458
.713	-.046	.835	-.267	.864	-.190	.863	-.384	.784	-.371	.861	-.285
.854	-.108	.919	-.116	.926	-.091	.923	-.138	.856	-.466	.918	-.051
.980	-.152	.987	.022	.975	-.028	.977	-.003	.926	-.176	.972	.015
1.074	-.186							.977	-.045		
1.122	-.067										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.065	-.022	.120	.024	-.116	.025	-.253	.019	-.322	.020	-.266
-.616	.004	.038	-.159	.075	-.348	0.000	0.000	.066	-.542	.076	-.569
0.000	0.000	.101	-.210	.297	-.329	.130	-.221	.136	-.434	.136	-.511
-.329	-.076	.185	-.256	.400	-.058	.298	-.290	.214	-.303	.221	-.436
-.172	-.016	.398	-.323	.604	-.156	.397	-.190	.292	-.251	.295	-.252
-.030	-.142	.737	.114	.785	.182	.501	-.239	.403	-.243	.396	-.365
.128	-.260			.967	.232	.603	.016	.489	-.142	.497	-.281
.418	-.222			1.000	.028	.703	.087	.594	-.203	.597	-.056
.564	-.171					.784	.135	.700	.039	.702	.074
.710	.025					.868	.195	.786	.146	.786	.131
.976	.245					.923	.214	.858	.204	.864	.172
1.072	.243							.919	.238	.912	.178
1.110	.183							.967	.141		
0.000	0.000										
CN=	.2322		.1683		.1876		.1838		.1102		.0604
CM=	-.0224		-.0551		-.0756		-.0879		-.0967		-.0934

$\alpha = 1.94^{\circ}$; $C_L = 0.247$

UPPER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	-.006	-.021	-.415	.023	-.461	.025	-.478	.022	-.301	.018	-.150
-.567	-.114	.035	-.658	.068	-.626	.079	-.547	.075	-.515	.077	-.392
-.452	-.187	.105	-.402	.134	-.521	.133	-.580	.129	-.466	.129	-.337
-.311	-.270	.178	-.437	.209	-.494	.214	-.485	.201	-.426	.209	-.288
-.023	-.367	.286	-.393	.294	-.482	.295	-.385	.294	-.219	.293	-.280
.133	-.284	.396	-.370	.404	-.401	.407	-.217	.397	-.217	.494	-.330
.272	-.229	.514	-.270	.497	-.259	.502	-.268	.495	-.292	.590	-.316
.416	-.207	.618	-.265	.599	-.305	.601	-.323	.594	-.337	.693	-.400
.565	-.117	.733	-.264	.700	-.363	.698	-.396	.693	-.417	.777	-.505
.713	-.076	.835	-.292	.864	-.223	.863	-.287	.784	-.500	.861	-.124
.854	-.127	.919	-.159	.926	-.081	.923	-.122	.856	-.435	.918	-.009
.980	-.176	.987	-.010	.975	-.050	.977	-.038	.926	-.164	.972	.032
1.074	-.217							.977	-.071		
1.122	-.107										

LOWER SURFACE

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
-.660	.085	-.022	.171	.024	.036	.025	-.079	.019	-.127	.020	-.032
-.616	.037	.038	-.081	.075	-.165	0.000	0.000	.066	-.332	.076	-.336
0.000	0.000	.101	-.144	.297	-.175	.130	-.233	.136	-.297	.136	-.314
-.329	-.047	.185	-.211	.400	-.158	.298	-.202	.214	-.257	.221	-.285
-.172	-.004	.398	-.082	.604	-.147	.397	-.229	.292	-.210	.295	-.301
-.030	-.116	.737	.116	.785	.172	.501	-.217	.403	-.210	.396	-.248
.128	-.238			.967	.208	.603	.021	.489	-.099	.497	-.242
.418	-.192			1.000	-.013	.703	.075	.594	-.215	.597	-.054
.564	-.097					.784	.138	.700	.052	.702	.093
.710	.076					.868	.209	.786	.170	.786	.171
.976	.267					.923	.238	.858	.237	.864	.216
1.072	.254							.919	.261	.912	.220
1.110	.181							.967	.149		
0.000	0.000										
CN=	.3236		.3192		.3094		.2797		.2544		.1958
CM=	-.0333		-.0677		-.0727		-.0775		-.1052		-.0890

[redacted]

TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(h) M = 0.99. Continued.

$\alpha = 2.42^o$; $C_L = 0.308$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	-.000	-.021	-.498	.023	-.533	.025	-.518	.022	-.373	.018	-.285
-.567	-.116	.035	-.693	.068	-.694	.079	-.612	.075	-.567	.077	-.536
-.452	-.195	.105	-.473	.134	-.611	.133	-.641	.129	-.570	.129	-.542
-.311	-.270	.178	-.451	.209	-.542	.214	-.591	.201	-.563	.209	-.454
-.023	-.371	.286	-.409	.294	-.464	.295	-.536	.294	-.494	.293	-.412
.133	-.287	.396	-.405	.404	-.435	.407	-.422	.397	-.367	.494	-.301
.272	-.243	.514	-.297	.497	-.266	.502	-.264	.495	-.251	.590	-.353
.416	-.222	.618	-.279	.599	-.310	.601	-.318	.594	-.312	.693	-.395
.565	-.134	.733	-.277	.700	-.386	.698	-.392	.693	-.407	.777	-.472
.713	-.089	.835	-.300	.864	-.220	.863	-.309	.784	-.502	.861	-.165
.654	-.133	.919	-.165	.926	-.093	.923	-.128	.856	-.499	.918	-.008
.980	-.178	.987	-.023	.975	-.060	.977	-.045	.926	-.154	.972	.021
1.074	-.232							.977	-.082		
1.122	-.117										
LOWER SURFACE											
-.650	.101	-.022	.213	.024	.101	.025	-.013	.019	-.082	.020	-.003
-.616	.051	.038	-.021	.075	-.102	0.000	0.000	.066	-.285	.076	-.290
0.000	0.000	.101	-.122	.297	-.159	.130	-.199	.136	-.263	.136	-.255
-.329	-.032	.185	-.168	.400	-.155	.298	-.166	.214	-.223	.221	-.270
-.172	.008	.398	-.094	.604	-.138	.397	-.237	.292	-.182	.295	-.296
-.030	-.102	.737	.126	.785	.170	.501	-.206	.403	-.207	.396	-.278
.128	-.225			.967	.209	.603	.020	.489	-.099	.497	-.260
.418	-.175			1.000	-.030	.703	.072	.594	-.224	.597	-.093
.564	-.058					.784	.137	.700	.044	.702	.093
.710	.097					.868	.214	.786	.167	.786	.172
.976	.283					.923	.251	.858	.235	.864	.212
1.072	.255							.919	.259	.912	.223
1.110	.183							.967	.144		
0.000	0.000										
CN=	.3639		.3631		.3496		.3510		.3280		.2570
CM=	-.0389		-.0686		-.0716		-.0803		-.1043		-.0820

$\alpha = 2.93^o$; $C_L = 0.376$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	-.004	-.021	-.579	.023	-.592	.025	-.570	.022	-.421	.018	-.364
-.567	-.135	.035	-.716	.068	-.738	.079	-.673	.075	-.623	.077	-.589
-.452	-.207	.105	-.561	.134	-.716	.133	-.690	.129	-.617	.129	-.617
-.311	-.280	.178	-.490	.209	-.669	.214	-.678	.201	-.637	.209	-.588
-.023	-.332	.286	-.444	.294	-.588	.295	-.619	.294	-.590	.293	-.551
.133	-.299	.395	-.430	.404	-.469	.407	-.610	.397	-.573	.494	-.382
.272	-.263	.514	-.322	.497	-.333	.502	-.331	.495	-.529	.590	-.402
.416	-.242	.618	-.303	.599	-.314	.601	-.312	.594	-.300	.693	-.391
.565	-.151	.733	-.291	.700	-.418	.698	-.401	.693	-.394	.777	-.485
.713	-.093	.835	-.308	.864	-.234	.863	-.273	.784	-.485	.861	-.082
.854	-.143	.919	-.162	.926	-.101	.923	-.106	.856	-.453	.918	-.009
.980	-.191	.987	-.035	.975	-.071	.977	-.062	.926	-.154	.972	.039
1.074	-.240							.977	-.082		
1.122	-.138										
LOWER SURFACE											
-.660	.112	-.022	.249	.024	.169	.025	.054	.019	-.036	.020	.048
-.616	.069	.038	.017	.075	-.046	0.000	0.000	.066	-.247	.076	-.200
C.000	0.000	.101	-.051	.297	-.140	.130	-.135	.136	-.205	.136	-.236
-.329	-.023	.185	-.034	.400	-.138	.298	-.188	.214	-.129	.221	-.244
-.172	.019	.398	-.086	.604	-.121	.397	-.240	.292	-.151	.295	-.275
-.030	-.096	.737	.130	.785	.163	.501	-.193	.403	-.218	.396	-.212
.128	-.215			.967	.199	.603	.031	.489	-.096	.497	-.253
.418	-.125			1.000	-.046	.703	.068	.594	-.211	.597	-.117
.564	-.033					.784	.130	.700	.051	.702	.046
.710	.112					.868	.215	.786	.163	.786	.138
.976	.290					.923	.260	.858	.230	.864	.188
1.072	.260							.919	.254	.912	.199
1.110	.130							.967	.135		
C.000	0.000										
CN=	.4144		.4210		.4190		.4101		.4113		.3165
CM=	-.0416		-.0684		-.0739		-.0794		-.1072		-.0736

[redacted]

TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = 3.43^{\circ}$; $C_L = 0.440$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	.000	-.021	-.351	.023	-.655	.025	-.613	.022	-.493	.018	-.413
	-.567	-.135	.035	-.748	.068	-.792	.079	-.722	.075	-.678	.077	-.632
	-.452	-.201	.105	-.695	.134	-.768	.133	-.735	.129	-.678	.129	-.650
	-.311	-.284	.178	-.530	.209	-.731	.214	-.722	.201	-.697	.209	-.632
	-.023	-.389	.286	-.466	.294	-.671	.295	-.676	.294	-.639	.293	-.598
	.133	-.319	.396	-.465	.404	-.588	.407	-.658	.397	-.644	.494	-.581
	.272	-.283	.514	-.346	.497	-.427	.502	-.633	.495	-.591	.590	-.576
	.416	-.269	.618	-.321	.599	-.314	.601	-.352	.594	-.596	.693	-.597
	.565	-.163	.733	-.307	.700	-.437	.698	-.393	.693	-.457	.777	-.229
	.713	-.103	.835	-.314	.864	-.280	.863	-.232	.784	-.459	.861	-.109
	.854	-.150	.919	-.159	.926	-.111	.923	-.111	.856	-.317	.918	-.061
	.980	-.201	.987	-.043	.975	-.077	.977	-.076	.926	-.148	.972	-.049
	1.074	-.256							.977	-.094		
	1.122	-.149										
						LOWER SURFACE						
	-.660	.124	-.022	.305	.024	.212	.025	.137	.019	.056	.020	.120
	-.616	.088	.038	.080	.075	-.020	0.000	0.000	.066	-.146	.076	-.157
	0.000	0.000	.101	-.006	.297	-.106	.130	-.092	.136	-.126	.136	-.218
	-.329	-.002	.185	-.057	.400	-.110	.298	-.179	.214	-.140	.221	-.237
	-.172	.023	.398	-.066	.604	-.045	.397	-.237	.292	-.162	.295	-.272
	-.030	-.085	.737	.134	.785	.166	.501	-.172	.403	-.217	.396	-.209
	.128	-.208			.967	.198	.603	.036	.489	-.085	.497	-.242
	.418	-.079			1.000	-.053	.703	.056	.594	-.197	.597	-.129
	.564	-.002					.784	.123	.700	.040	.702	.038
	.710	.124					.868	.218	.786	.159	.786	.113
	.976	.299					.923	.269	.858	.230	.864	.155
	1.072	.265							.919	.251	.912	.175
	1.110	.183							.967	.128		
	0.000	0.000										
CN=	.4592		.4772		.5025		.4759		.4827		.3885	
CM=	-.0473		-.0683		-.0870		-.0835		-.1110		-.0834	

$\alpha = 3.94^{\circ}$; $C_L = 0.498$

	STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
						UPPER SURFACE						
	-.660	-.007	-.021	-.707	.023	-.708	.025	-.666	.022	-.538	.018	-.454
	-.567	-.143	.035	-.808	.068	-.834	.079	-.757	.075	-.716	.077	-.676
	-.452	-.191	.105	-.773	.134	-.814	.133	-.768	.129	-.722	.129	-.696
	-.311	-.296	.178	-.557	.209	-.789	.214	-.763	.201	-.729	.209	-.679
	-.023	-.405	.286	-.476	.294	-.772	.295	-.730	.294	-.681	.293	-.638
	.133	-.333	.396	-.490	.404	-.701	.407	-.705	.397	-.682	.494	-.643
	.272	-.300	.514	-.366	.497	-.478	.502	-.697	.495	-.630	.590	-.617
	.416	-.283	.618	-.330	.599	-.336	.601	-.660	.594	-.640	.693	-.648
	.565	-.160	.733	-.320	.700	-.445	.698	-.428	.693	-.669	.777	-.221
	.713	-.105	.835	-.320	.864	-.304	.863	-.203	.784	-.488	.861	-.155
	.854	-.159	.919	-.157	.926	-.121	.923	-.118	.856	-.263	.918	-.123
	.980	-.206	.987	-.053	.975	-.097	.977	-.089	.926	-.195	.972	-.122
	1.074	-.262							.977	-.207		
	1.122	-.164										
						LOWER SURFACE						
	-.660	.133	-.022	.332	.024	.260	.025	.159	.019	.097	.020	.163
	-.616	.101	.038	.118	.075	.035	0.000	0.000	.066	-.113	.076	-.118
	0.000	0.000	.101	.031	.297	-.086	.130	-.067	.136	-.113	.136	-.201
	-.329	.017	.185	-.017	.400	-.087	.298	-.166	.214	-.122	.221	-.229
	-.172	.038	.398	-.047	.604	-.043	.397	-.196	.292	-.151	.295	-.256
	-.030	-.075	.737	.140	.785	.169	.501	-.162	.403	-.200	.396	-.208
	.128	-.201			.967	.198	.603	.024	.489	-.063	.497	-.259
	.418	-.061			1.000	-.063	.703	.051	.594	-.207	.597	-.169
	.564	.005					.784	.119	.700	.029	.702	-.010
	.710	.137					.868	.218	.786	.140	.786	.075
	.976	.304					.923	.275	.858	.214	.864	.125
	1.072	.263							.919	.232	.912	.132
	1.110	.185							.967	.086		
	0.000	0.000										
CN=	.4917		.5229		.5664		.5458		.5382		.4212	
CM=	-.0459		-.0687		-.0922		-.0957		-.1219		-.0840	

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = 4.95^{\circ}$; $C_L = 0.612$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.017	-.021	-.823	.023	-.798	.025	-.766	.022	-.658	.018	-.568
-.567	-.089	.035	-.963	.068	-.917	.079	-.829	.075	-.802	.077	-.757
-.452	-.228	.105	-.925	.134	-.901	.133	-.853	.129	-.799	.129	-.776
-.311	-.319	.178	-.871	.209	-.877	.214	-.840	.201	-.807	.209	-.755
-.023	-.436	.286	-.492	.294	-.849	.295	-.806	.294	-.764	.293	-.728
.133	-.363	.396	-.514	.404	-.813	.407	-.788	.397	-.765	.494	-.713
.272	-.315	.514	-.406	.497	-.789	.502	-.780	.495	-.730	.590	-.700
.416	-.282	.618	-.360	.599	-.428	.601	-.772	.594	-.722	.693	-.720
.565	-.170	.733	-.341	.700	-.460	.698	-.796	.693	-.506	.777	-.680
.713	-.123	.835	-.328	.864	-.248	.863	-.352	.784	-.309	.861	-.283
.854	-.178	.919	-.159	.926	-.116	.923	-.298	.856	-.293	.918	-.255
.980	-.225	.987	-.061	.975	-.097	.977	-.286	.926	-.283	.972	-.225
1.074	-.294							.977	-.281		
1.122	-.199										
				LOWER SURFACE							
-.660	.157	-.022	.391	.024	.349	.025	.255	.019	.205	.020	.235
-.616	.139	.038	.199	.075	.112	0.000	0.000	.066	-.019	.076	-.059
0.000	0.000	.101	.104	.297	-.031	.130	.002	.136	-.059	.136	-.154
-.329	.038	.185	.045	.400	-.055	.298	-.099	.214	-.088	.221	-.184
-.172	.057	.398	.001	.604	-.039	.397	-.085	.292	-.115	.295	-.228
-.030	-.054	.737	.158	.785	.183	.501	-.169	.403	-.054	.396	-.273
.128	-.175			.967	.195	.603	.039	.489	-.082	.497	-.256
.418	-.016			1.000	-.067	.703	.053	.594	-.217	.597	-.206
.564	.043					.784	.118	.700	-.005	.702	-.047
.710	.168					.868	.219	.786	.114	.786	.037
.976	.327					.923	.282	.858	.189	.864	.089
1.072	.283							.919	.211	.912	.083
1.110	.190							.967	.043		
0.000	0.000										
CN=	.5673		.6404		.6737		.7250		.5963		.5331
CM=	-.0532		-.0688		-.0988		-.1465		-.1111		-.1125

$\alpha = 5.96^{\circ}$; $C_L = 0.713$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	.001	-.021	-.907	.023	-.882	.025	-.851	.022	-.760	.018	-.676
-.567	-.186	.035	-1.028	.068	-.988	.079	-.916	.075	-.873	.077	-.834
-.452	-.263	.105	-1.007	.134	-.957	.133	-.917	.129	-.863	.129	-.856
-.311	-.358	.178	-1.003	.209	-.942	.214	-.900	.201	-.866	.209	-.832
-.023	-.457	.286	-.580	.294	-.917	.295	-.871	.294	-.829	.293	-.799
.133	-.378	.396	-.531	.404	-.911	.407	-.856	.397	-.828	.494	-.795
.272	-.327	.514	-.438	.497	-.887	.502	-.843	.495	-.807	.590	-.777
.416	-.293	.618	-.385	.599	-.730	.601	-.838	.594	-.804	.693	-.773
.565	-.184	.733	-.360	.700	-.486	.698	-.853	.693	-.454	.777	-.620
.713	-.140	.835	-.331	.864	-.276	.863	-.406	.784	-.369	.861	-.348
.854	-.201	.919	-.164	.926	-.149	.923	-.391	.856	-.344	.918	-.332
.980	-.249	.987	-.080	.975	-.105	.977	-.374	.926	-.338	.972	-.309
1.074	-.321							.977	-.336		
1.122	-.232										
				LOWER SURFACE							
-.660	.175	-.022	.428	.024	.403	.025	.333	.019	.279	.020	.283
-.616	.173	.038	.243	.075	.180	0.000	0.000	.066	.042	.076	-.004
0.000	0.000	.101	.166	.297	.007	.130	.053	.136	-.017	.136	-.105
-.329	.066	.185	.103	.400	-.020	.298	-.013	.214	-.049	.221	-.156
-.172	.072	.398	.038	.604	-.031	.397	-.063	.292	-.088	.295	-.197
-.030	-.021	.737	.172	.785	.200	.501	-.143	.403	-.009	.396	-.259
.128	-.114			.967	.189	.603	.044	.489	-.078	.497	-.258
.418	.023			1.000	-.098	.703	.052	.594	-.217	.597	-.213
.564	.070					.784	.114	.700	-.018	.702	-.063
.710	.190					.868	.223	.786	.109	.786	.022
.976	.340					.923	.280	.858	.181	.864	.072
1.072	.289							.919	.202	.912	.073
1.110	.198							.967	.027		
0.000	0.000										
CN=	.6601		.7248		.7835		.8221		.6684		.6072
CM=	-.0441		-.0719		-.1171		-.1624		-.1182		-.1200

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = 6.96^{\circ}$; $C_L = 0.807$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.044	-.021	-.984	.023	-.957	.025	-.921	.022	-.827	.018	-.755
-.567	-.228	.035	-1.110	.068	-1.050	.079	-.972	.075	-.948	.077	-.903
-.452	-.286	.105	-1.099	.134	-1.019	.133	-.976	.129	-.935	.129	-.910
-.311	-.371	.178	-1.072	.209	-1.003	.214	-.956	.201	-.937	.209	-.886
-.023	-.470	.286	-.805	.294	-.984	.295	-.925	.294	-.890	.293	-.860
.133	-.389	.396	-.596	.404	-.965	.407	-.916	.397	-.876	.494	-.849
.272	-.336	.514	-.461	.497	-.955	.502	-.907	.495	-.865	.590	-.836
.416	-.290	.618	-.411	.599	-.912	.601	-.898	.594	-.849	.693	-.837
.565	-.199	.733	-.378	.700	-.682	.698	-.509	.693	-.554	.777	-.738
.713	-.159	.835	-.342	.864	-.355	.863	-.430	.784	-.450	.861	-.417
.854	-.219	.919	-.176	.926	-.200	.923	-.428	.856	-.395	.918	-.389
.980	-.273	.987	-.094	.975	-.149	.977	-.422	.926	-.354	.972	-.396
1.074	-.342							.977	-.352		
1.122	-.273										
					LOWER SURFACE						
-.660	.190	-.022	.466	.024	.467	.025	.400	.019	.327	.020	.318
-.616	.190	.038	.317	.075	.246	0.000	0.000	.066	.109	.076	.033
0.000	0.000	.101	.224	.297	.055	.130	.101	.136	.025	.136	-.064
-.329	.096	.185	.149	.400	.010	.298	.008	.214	-.021	.221	-.126
-.172	.102	.398	.075	.604	-.023	.397	-.052	.292	-.069	.295	-.184
-.030	.007	.737	.183	.785	.190	.501	-.127	.403	-.012	.396	-.222
.128	-.052			.967	.161	.603	.034	.489	-.080	.497	-.256
.418	.063			1.000	-.163	.703	.034	.594	-.214	.597	-.220
.564	.109					.784	.075	.700	-.024	.702	-.073
.710	.213					.868	.210	.786	.101	.786	.010
.976	.354					.923	.271	.858	.175	.864	.056
1.072	.298							.919	.197	.912	.053
1.110	.198							.967	.013		
0.000	0.000										
CN=	.7419		.8234		.8975		.8328		.7389		.6826
CM=	-.0411		-.0761		-.1409		-.1450		-.1279		-.1348

$\alpha = 7.99^{\circ}$; $C_L = 0.893$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.092	-.021	-1.042	.023	-1.022	.025	-.974	.022	-.899	.018	-.820
-.567	-.259	.035	-1.169	.068	-1.100	.079	-1.027	.075	-1.006	.077	-.959
-.452	-.319	.105	-1.149	.134	-1.084	.133	-1.026	.129	-.988	.129	-.976
-.311	-.393	.178	-1.107	.209	-1.061	.214	-1.014	.201	-.985	.209	-.947
-.023	-.487	.286	-.954	.294	-1.027	.295	-.978	.294	-.949	.293	-.922
.133	-.402	.396	-.782	.404	-1.010	.407	-.976	.397	-.932	.494	-.899
.272	-.347	.514	-.577	.497	-.968	.502	-.962	.495	-.920	.590	-.889
.416	-.305	.618	-.449	.599	-.800	.601	-.921	.594	-.902	.693	-.872
.565	-.226	.733	-.401	.700	-.621	.698	-.625	.693	-.714	.777	-.766
.713	-.185	.835	-.373	.864	-.384	.863	-.511	.784	-.548	.861	-.493
.854	-.244	.919	-.239	.926	-.313	.923	-.495	.856	-.463	.918	-.447
.980	-.302	.987	-.112	.975	-.284	.977	-.474	.926	-.408	.972	-.459
1.074	-.375							.977	-.413		
1.122	-.310										
					LOWER SURFACE						
-.660	.197	-.022	.490	.024	.512	.025	.447	.019	.369	.020	.364
-.616	.217	.038	.357	.075	.302	0.000	0.000	.066	.141	.076	.073
0.000	0.000	.101	.266	.297	.095	.130	.141	.136	.054	.136	-.027
-.329	.130	.185	.202	.400	.043	.298	.035	.214	.001	.221	-.101
-.172	.128	.398	.110	.604	-.012	.397	-.034	.292	-.043	.295	-.165
-.030	.043	.737	.199	.785	.183	.501	-.114	.403	-.008	.396	-.154
.128	-.005			.967	.121	.603	.029	.489	-.076	.497	-.241
.418	.102			1.000	-.293	.703	.019	.594	-.207	.597	-.225
.564	.137					.784	.057	.700	-.018	.702	-.085
.710	.238					.868	.210	.786	.100	.786	-.000
.976	.368					.923	.270	.858	.182	.864	.049
1.072	.295							.919	.193	.912	.045
1.110	.201							.967	.009		
0.000	0.000										
CN=	.8307		.9363		.9365		.9087		.8211		.7527
CM=	-.0383		-.0922		-.1397		-.1611		-.1480		-.1456

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Continued.

$\alpha = 9.15^{\circ}$; $C_L = 0.982$

	STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE							
	-.660	-.143	-.021	-1.087	.023	-1.089	.025	-1.047	.022	-.960	.018	-.899
	-.567	-.332	.035	-1.211	.068	-1.159	.079	-1.092	.075	-1.060	.077	-1.015
	-.452	-.353	.105	-1.166	.134	-1.132	.133	-1.082	.129	-1.043	.129	-1.021
	-.311	-.423	.178	-1.165	.209	-1.100	.214	-1.062	.201	-1.036	.209	-.996
	-.023	-.495	.286	-.978	.294	-1.076	.295	-1.038	.294	-1.002	.293	-.968
	.133	-.412	.396	-.845	.404	-1.033	.407	-1.032	.397	-.984	.494	-.942
	.272	-.357	.514	-.794	.497	-.888	.502	-1.005	.495	-.967	.590	-.929
	.416	-.316	.618	-.716	.599	-.722	.601	-.942	.594	-.938	.693	-.910
	.565	-.246	.733	-.590	.700	-.634	.698	-.810	.693	-.831	.777	-.826
	.713	-.212	.835	-.468	.864	-.464	.863	-.599	.784	-.624	.861	-.544
	.854	-.231	.919	-.323	.926	-.448	.923	-.520	.856	-.550	.918	-.506
	.980	-.325	.987	-.171	.975	-.411	.977	-.434	.926	-.519	.972	-.509
	1.074	-.392							.977	-.534		
	1.122	-.417										
					LOWER SURFACE							
	-.660	.204	-.022	.513	.024	.556	.025	.479	.019	.409	.020	.400
	-.616	.247	.038	.401	.075	.353	0.000	0.000	.066	.188	.076	.120
	0.000	0.000	.101	.317	.297	.126	.130	.183	.136	.095	.136	-.003
	-.329	.166	.185	.245	.400	.080	.298	.063	.214	.032	.221	-.071
	-.172	.171	.398	.156	.604	.002	.397	-.014	.292	-.007	.295	-.078
	-.030	.095	.737	.209	.785	.173	.501	-.101	.403	.002	.396	-.147
	.128	.048			.967	.099	.603	.031	.489	-.067	.497	-.235
	.418	.135			1.000	-.368	.703	.016	.594	-.194	.597	-.221
	.564	.170					.784	.058	.700	-.002	.702	-.082
	.710	.269					.868	.219	.786	.108	.786	-.003
	.976	.386					.923	.282	.858	.186	.864	.049
	1.072	.317							.919	.193	.912	.042
	1.110	.208							.967	.006		
	0.000	0.000										
CN=		.9350		1.0832		.9836		.9916		.9080		.8193
CM=		-.0340		-.1326		-.1474		-.1788		-.1711		-.1568

$\alpha = 10.18^{\circ}$; $C_L = 1.056$

	STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE							
	-.660	-.199	-.021	-1.146	.023	-1.122	.025	-1.091	.022	-1.021	.018	-.957
	-.567	-.377	.035	-1.206	.068	-1.195	.079	-1.129	.075	-1.106	.077	-1.064
	-.452	-.385	.105	-1.193	.134	-1.152	.133	-1.126	.129	-1.089	.129	-1.072
	-.311	-.449	.178	-1.133	.209	-1.132	.214	-1.105	.201	-1.086	.209	-1.041
	-.023	-.511	.286	-1.003	.294	-1.095	.295	-1.061	.294	-1.044	.293	-1.019
	.133	-.418	.396	-.968	.404	-.926	.407	-1.040	.397	-1.020	.494	-.990
	.272	-.367	.514	-.906	.497	-.817	.502	-1.018	.495	-.993	.590	-.951
	.416	-.334	.618	-.868	.599	-.718	.601	-.934	.594	-.972	.693	-.890
	.565	-.257	.733	-.809	.700	-.625	.698	-.816	.693	-.873	.777	-.727
	.713	-.244	.835	-.656	.864	-.497	.863	-.654	.784	-.764	.861	-.589
	.854	-.304	.919	-.344	.926	-.482	.923	-.528	.856	-.735	.918	-.545
	.980	-.342	.987	-.180	.975	-.465	.977	-.425	.926	-.693	.972	-.545
	1.074	-.409							.977	-.648		
	1.122	-.469										
					LOWER SURFACE							
	-.660	.214	-.022	.518	.024	.577	.025	.526	.019	.450	.020	.421
	-.616	.271	.038	.448	.075	.397	0.000	0.000	.066	.234	.076	.163
	0.000	0.000	.101	.375	.297	.170	.130	.214	.136	.130	.136	.048
	-.329	.202	.185	.290	.400	.110	.298	.090	.214	.059	.221	-.019
	-.172	.183	.398	.181	.604	.013	.397	.009	.292	.020	.295	-.075
	-.030	.128	.737	.231	.785	.170	.501	-.070	.403	.014	.396	-.139
	.128	.088			.967	.080	.603	.036	.489	-.055	.497	-.213
	.418	.164			1.000	-.501	.703	.019	.594	-.185	.597	-.196
	.564	.203					.784	.047	.700	.011	.702	-.079
	.710	.293					.868	.214	.786	.115	.786	-.001
	.976	.400					.923	.278	.858	.192	.864	.045
	1.072	.324							.919	.199	.912	.031
	1.110	.206							.967	.005		
	0.000	0.000										
CN=		1.0169		1.2069		1.0001		1.0289		.9960		.8617
CM=		-.0292		-.1704		-.1466		-.1814		-.1998		-.1571

[redacted]

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99. Concluded.

$\alpha = 11.21^{\circ}$; $C_L = 1.107$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.232	-.021	-1.182	.023	-1.167	.025	-1.132	.022	-1.077	.018	-1.020
-.567	-.414	.035	-1.200	.068	-1.131	.079	-1.167	.075	-1.125	.077	-1.110
-.452	-.418	.105	-1.202	.134	-1.141	.133	-1.140	.129	-1.119	.129	-1.108
-.311	-.473	.178	-1.137	.209	-1.083	.214	-1.098	.201	-1.102	.209	-1.076
-.023	-.519	.286	-1.082	.294	-.968	.295	-1.067	.294	-1.076	.293	-1.040
.133	-.433	.396	-1.047	.404	-.874	.407	-.936	.397	-1.044	.494	-.989
.272	-.376	.514	-1.038	.497	-.793	.502	-.629	.495	-.965	.590	-.860
.416	-.350	.618	-1.066	.599	-.729	.601	-.497	.594	-.958	.693	-.715
.565	-.293	.733	-1.021	.700	-.654	.698	-.456	.693	-.847	.777	-.658
.713	-.267	.835	-.786	.864	-.634	.863	-.460	.784	-.709	.861	-.605
.854	-.329	.919	-.395	.926	-.619	.923	-.448	.856	-.629	.918	-.579
.980	-.369	.987	-.137	.975	-.556	.977	-.441	.926	-.625	.972	-.579
1.074	-.434							.977	-.590		
1.122	-.504										
					LOWER SURFACE						
-.660	.223	-.022	.531	.024	.603	.025	.567	.019	.489	.020	.461
-.616	.293	.038	.489	.075	.451	0.000	0.000	.066	.277	.076	.210
0.000	0.000	.101	.399	.297	.205	.130	.268	.136	.180	.136	.104
-.329	.227	.185	.333	.400	.151	.298	.123	.214	.109	.221	.009
-.172	.219	.398	.219	.604	.031	.397	.047	.292	.072	.295	-.034
-.030	.170	.737	.249	.785	.179	.501	-.011	.403	.039	.396	-.115
.128	.125			.967	.084	.603	.057	.489	-.038	.497	-.188
.418	.202			1.000	-.580	.703	.028	.594	-.164	.597	-.184
.564	.236					.784	.065	.700	.028	.702	-.073
.710	.315					.868	.225	.786	.122	.786	.002
.976	.419					.923	.290	.858	.200	.864	.044
1.072	.338							.919	.206	.912	.040
1.110	.219							.967	.010		
0.000	0.000										
CN=	1.1048		1.3315		1.0325		.9009		1.0095		.8718
CM=	-.0289		-.2061		-.1674		-.1208		-.1874		-.1469

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(i) M = 1.00.

$\alpha = -1.11^{\circ}$; $C_L = -0.142$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	-.027	-.021	-.013	.023	-.026	.025	.029	.022	.172	.018	.296
-.567	-.084	.035	-.213	.068	-.110	.079	-.089	.075	-.040	.077	.032
-.452	-.160	.105	-.123	.134	-.152	.133	-.098	.129	-.064	.129	.002
-.311	-.227	.178	-.149	.209	-.167	.214	-.104	.201	-.089	.209	-.033
-.023	-.312	.286	-.151	.294	-.129	.295	-.114	.294	-.089	.293	-.075
.133	-.228	.396	-.145	.404	-.123	.407	-.140	.397	-.095	.494	-.165
.272	-.164	.514	-.116	.497	-.159	.502	-.185	.495	-.134	.590	-.228
.416	-.031	.618	-.146	.599	-.186	.601	-.230	.594	-.153	.693	-.318
.565	.028	.733	-.155	.700	-.255	.698	-.297	.693	-.252	.777	-.407
.713	.024	.835	-.203	.864	-.181	.863	-.360	.784	-.324	.861	-.277
.854	-.033	.919	-.142	.926	-.071	.923	-.128	.856	-.324	.918	-.233
.980	-.074	.987	.018	.975	-.009	.977	.016	.926	-.183	.972	-.119
1.074	-.125							.977	-.009		
1.122	-.049										
LOWER SURFACE											
-.660	.010	-.022	-.022	.024	-.380	.025	-.568	.019	-.624	.020	-.700
-.616	-.083	.038	-.290	.075	-.490	0.000	0.000	.066	-.842	.076	-.713
0.000	0.000	.101	-.346	.297	-.523	.130	-.764	.136	-.837	.136	-.700
-.329	-.151	.185	-.339	.400	-.543	.298	-.699	.214	-.794	.221	-.597
-.172	-.042	.398	-.411	.604	-.099	.397	-.697	.292	-.785	.295	-.483
-.030	-.171	.737	.038	.785	.134	.501	-.223	.403	-.425	.396	-.395
.128	-.292			.967	.204	.603	.033	.489	-.302	.497	-.315
.418	-.248			1.000	.015	.703	.114	.594	-.252	.597	-.233
.564	-.253					.784	.160	.700	-.036	.702	-.187
.710	-.062					.868	.234	.786	.093	.786	.004
.976	.203					.923	.222	.858	.225	.864	-.086
1.072	.219							.919	.242	.912	-.055
1.110	.175							.967	.181		
0.000	0.000										
CN=	.0576		-.0675		-.0750		-.1106		-.1920		-.1923
CM=	.0055		-.0358		-.0628		-.0928		-.0840		-.0572

$\alpha = -0.11^{\circ}$; $C_L = -0.018$

STA .133 X/C	CP	STA .307 X/C	CP	STA .480 X/C	CP	STA .653 X/C	CP	STA .804 X/C	CP	STA .933 X/C	CP
UPPER SURFACE											
-.660	-.025	-.021	-.087	.023	-.125	.025	-.072	.022	.095	.018	.211
-.567	-.101	.035	-.340	.068	-.298	.079	-.187	.075	-.102	.077	-.043
-.452	-.180	.105	-.280	.134	-.237	.133	-.198	.129	-.123	.129	-.055
-.311	-.247	.178	-.322	.209	-.229	.214	-.190	.201	-.156	.209	-.094
-.023	-.326	.286	-.220	.294	-.213	.295	-.175	.294	-.143	.293	-.123
.133	-.251	.396	-.157	.404	-.180	.407	-.174	.397	-.150	.494	-.201
.272	-.190	.514	-.153	.497	-.192	.502	-.209	.495	-.203	.590	-.247
.416	-.126	.618	-.159	.599	-.255	.601	-.258	.594	-.189	.693	-.313
.565	.004	.733	-.179	.700	-.288	.698	-.334	.693	-.275	.777	-.412
.713	.004	.835	-.233	.864	-.269	.863	-.227	.784	-.300	.861	-.397
.854	-.049	.919	-.182	.926	-.061	.923	-.109	.856	-.337	.918	-.058
.980	-.099	.987	.012	.975	-.016	.977	.018	.926	-.186	.972	.022
1.074	-.146							.977	-.024		
1.122	-.069										
LOWER SURFACE											
-.660	.015	-.022	.063	.024	-.213	.025	-.446	.019	-.529	.020	-.531
-.616	-.055	.038	-.216	.075	-.412	0.000	0.000	.066	-.759	.076	-.717
0.000	0.000	.101	-.291	.297	-.451	.130	-.659	.136	-.713	.136	-.712
-.329	-.117	.185	-.290	.400	-.502	.298	-.581	.214	-.675	.221	-.727
-.172	-.033	.398	-.382	.604	-.101	.397	-.235	.292	-.571	.295	-.506
-.030	-.158	.737	.061	.785	.152	.501	-.187	.403	-.187	.396	-.312
.128	-.271			.967	.230	.603	.042	.489	-.138	.497	-.170
.418	-.238			1.000	.010	.703	.106	.594	-.155	.597	-.028
.564	-.227					.784	.149	.700	.032	.702	.059
.710	-.034					.868	.201	.786	.141	.786	.120
.976	.213					.923	.223	.858	.221	.864	.157
1.072	.224							.919	.248	.912	.161
1.110	.171							.967	.145		
0.000	0.000										
CN=	.1319		.0345		.0319		.0040		-.0473		-.0596
CM=	.0023		-.0405		-.0729		-.0871		-.0990		-.1018

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TABLE V – Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(i) M = 1.00. Continued.

$\alpha = 0.95^{\circ}$; $C_L = 0.114$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.029	-.021	-.276	.023	-.345	.025	-.278	.022	-.037	.018	.054
-.567	-.130	.035	-.514	.068	-.465	.079	-.379	.075	-.218	.077	-.201
-.452	-.197	.105	-.366	.134	-.416	.133	-.291	.129	-.230	.129	-.190
-.311	-.265	.178	-.382	.209	-.402	.214	-.238	.201	-.238	.209	-.200
-.023	-.351	.286	-.341	.294	-.334	.295	-.191	.294	-.212	.293	-.215
.133	-.270	.396	-.265	.404	-.173	.407	-.228	.397	-.187	.494	-.222
.272	-.214	.514	-.199	.497	-.197	.502	-.259	.495	-.239	.590	-.281
.416	-.166	.618	-.203	.599	-.279	.601	-.284	.594	-.290	.693	-.364
.565	-.060	.733	-.209	.700	-.320	.698	-.354	.693	-.380	.777	-.466
.713	-.023	.835	-.253	.864	-.332	.863	-.183	.784	-.267	.861	-.370
.854	-.074	.919	-.184	.926	-.082	.923	-.080	.856	-.297	.918	-.054
.980	-.127	.987	-.004	.975	-.051	.977	.003	.926	-.198	.972	.009
1.074	-.183							.977	-.037		
1.122	-.089										
					LOWER SURFACE						
-.660	.045	-.022	.131	.024	-.095	.025	-.306	.019	-.328	.020	-.294
-.616	-.015	.038	-.119	.075	-.314	0.000	0.000	.066	-.518	.076	-.539
0.000	0.000	.101	-.185	.297	-.391	.130	-.496	.136	-.427	.136	-.547
-.329	-.071	.185	-.234	.400	-.170	.298	-.208	.214	-.358	.221	-.481
-.172	-.015	.398	-.308	.604	-.122	.397	-.199	.292	-.267	.295	-.320
-.030	-.126	.737	.098	.785	.177	.501	-.171	.403	-.226	.396	-.332
.128	-.250			.967	.232	.603	.004	.489	-.149	.497	-.246
.418	-.214			1.000	-.003	.703	.084	.594	-.216	.597	-.029
.564	-.184					.784	.143	.700	.040	.702	.091
.710	.009					.868	.206	.786	.152	.786	.153
.976	.237					.923	.224	.858	.218	.864	.181
1.072	.235							.919	.249	.912	.188
1.110	.174							.967	.156		
0.000	0.000										
CN=	.2294		.1722		.1856		.1290		.0967		.0718
CM=	-.0039		-.0500		-.0823		-.0787		-.0942		-.1012

$\alpha = 1.96^{\circ}$; $C_L = 0.237$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.040	-.021	-.463	.023	-.460	.025	-.436	.022	-.268	.018	-.157
-.567	-.145	.035	-.650	.068	-.637	.079	-.528	.075	-.491	.077	-.358
-.452	-.218	.105	-.466	.134	-.535	.133	-.565	.129	-.455	.129	-.357
-.311	-.280	.178	-.428	.209	-.502	.214	-.481	.201	-.410	.209	-.281
-.023	-.362	.286	-.383	.294	-.461	.295	-.395	.294	-.255	.293	-.268
.133	-.286	.396	-.352	.404	-.318	.407	-.232	.397	-.226	.494	-.318
.272	-.235	.514	-.235	.497	-.200	.502	-.277	.495	-.290	.590	-.362
.416	-.208	.618	-.218	.599	-.282	.601	-.317	.594	-.326	.693	-.440
.565	-.094	.733	-.232	.700	-.348	.698	-.388	.693	-.412	.777	-.536
.713	-.046	.835	-.278	.864	-.359	.863	-.266	.784	-.494	.861	-.174
.854	-.106	.919	-.182	.926	-.108	.923	-.101	.856	-.353	.918	-.053
.980	-.149	.987	-.022	.975	-.075	.977	-.052	.926	-.130	.972	.002
1.074	-.206							.977	-.068		
1.122	-.110										
					LOWER SURFACE						
-.660	.059	-.022	.185	.024	-.000	.025	-.128	.019	-.145	.020	-.056
-.616	.013	.038	-.060	.075	-.186	0.000	0.000	.066	-.348	.076	-.321
0.000	0.000	.101	-.137	.297	-.173	.130	-.224	.136	-.341	.136	-.394
-.329	-.050	.185	-.195	.400	-.212	.298	-.211	.214	-.257	.221	-.302
-.172	.000	.398	-.154	.604	-.134	.397	-.229	.292	-.256	.295	-.272
-.030	-.116	.737	.117	.785	.167	.501	-.202	.403	-.201	.396	-.302
.128	-.228			.967	.206	.603	.008	.489	-.096	.497	-.258
.418	-.194			1.000	-.044	.703	.068	.594	-.201	.597	-.097
.564	-.114					.784	.130	.700	.045	.702	.073
.710	.071					.868	.201	.786	.165	.786	.150
.976	.271					.923	.227	.858	.233	.864	.197
1.072	.250							.919	.255	.912	.211
1.110	.182							.967	.139		
0.000	0.000										
CN=	.3181		.2964		.2976		.2681		.2346		.1874
CM=	-.0186		-.0590		-.0790		-.0755		-.1002		-.0931

[redacted]

TABLE V — *Continued*

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(i) M = 1.00. Continued.

$\alpha = 2.45^{\circ}$; $C_L = 0.299$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.043	-.021	-.517	.023	-.516	.025	-.504	.022	-.345	.018	-.262
-.567	-.161	.035	-.688	.068	-.685	.079	-.602	.075	-.557	.077	-.514
-.452	-.220	.105	-.580	.134	-.634	.133	-.616	.129	-.557	.129	-.543
-.311	-.292	.178	-.461	.209	-.536	.214	-.572	.201	-.555	.209	-.438
-.023	-.369	.286	-.405	.294	-.529	.295	-.521	.294	-.491	.293	-.320
.133	-.292	.396	-.391	.404	-.423	.407	-.178	.397	-.388	.494	-.280
.272	-.247	.514	-.255	.497	-.225	.502	-.263	.495	-.239	.590	-.342
.416	-.224	.618	-.242	.599	-.291	.601	-.310	.594	-.310	.693	-.407
.565	-.127	.733	-.251	.700	-.372	.698	-.395	.693	-.403	.777	-.530
.713	-.060	.835	-.289	.864	-.373	.863	-.332	.784	-.492	.861	-.207
.854	-.114	.919	-.185	.926	-.114	.923	-.123	.856	-.468	.918	-.060
.980	-.159	.987	-.029	.975	-.087	.977	-.084	.926	-.134	.972	-.015
1.074	-.215							.977	-.070		
1.122	-.122										
					LOWER SURFACE						
-.660	.074	-.022	.228	.024	.115	.025	-.011	.019	-.068	.020	-.002
-.616	.027	.038	-.014	.075	-.111	0.000	0.000	.066	-.291	.076	-.305
0.000	0.000	.101	-.103	.297	-.141	.130	-.188	.136	-.290	.136	-.362
-.329	-.038	.185	-.160	.400	-.142	.298	-.160	.214	-.238	.221	-.253
-.172	.010	.398	-.079	.604	-.128	.397	-.217	.292	-.220	.295	-.265
-.030	-.102	.737	.131	.785	.171	.501	-.198	.403	-.205	.396	-.303
.128	-.218			.967	.196	.603	-.000	.489	-.102	.497	-.253
.418	-.156			1.000	-.057	.703	.061	.594	-.212	.597	-.121
.564	-.078					.784	.114	.700	.035	.702	.069
.710	.101					.868	.203	.786	.162	.786	.149
.976	.288					.923	.239	.858	.230	.864	.189
1.072	.257							.919	.250	.912	.199
1.110	.185							.967	.138		
0.000	0.000										
CN=	.3718		.3686		.3718		.3226		.3120		.2324
CM=	-.0249		-.0648		-.0820		-.0767		-.1005		-.0843

$\alpha = 2.96^{\circ}$; $C_L = 0.363$

STA X/C	.133 CP	STA X/C	.307 CP	STA X/C	.480 CP	STA X/C	.653 CP	STA X/C	.804 CP	STA X/C	.933 CP
					UPPER SURFACE						
-.660	-.051	-.021	-.612	.023	-.569	.025	-.554	.022	-.412	.018	-.332
-.567	-.171	.035	-.712	.068	-.734	.079	-.652	.075	-.622	.077	-.570
-.452	-.236	.105	-.622	.134	-.691	.133	-.668	.129	-.619	.129	-.598
-.311	-.295	.178	-.485	.209	-.651	.214	-.665	.201	-.623	.209	-.559
-.023	-.379	.286	-.424	.294	-.599	.295	-.623	.294	-.570	.293	-.534
.133	-.306	.396	-.406	.404	-.418	.407	-.568	.397	-.575	.494	-.514
.272	-.258	.514	-.279	.497	-.259	.502	-.277	.495	-.290	.590	-.472
.416	-.230	.618	-.261	.599	-.284	.601	-.312	.594	-.308	.693	-.372
.565	-.127	.733	-.267	.700	-.402	.698	-.401	.693	-.396	.777	-.487
.713	-.073	.835	-.302	.864	-.390	.863	-.320	.784	-.492	.861	-.173
.854	-.120	.919	-.184	.926	-.134	.923	-.136	.856	-.388	.918	-.051
.980	-.178	.987	-.041	.975	-.101	.977	-.103	.926	-.137	.972	-.003
1.074	-.234							.977	-.094		
1.122	-.142										
					LOWER SURFACE						
-.660	.077	-.022	.263	.024	.165	.025	.063	.019	-.031	.020	.040
-.616	.042	.038	.019	.075	-.054	0.000	0.000	.066	-.251	.076	-.258
0.000	0.000	.101	-.078	.297	-.138	.130	-.156	.136	-.255	.136	-.258
-.329	-.022	.185	-.088	.400	-.129	.298	-.176	.214	-.187	.221	-.231
-.172	.023	.398	-.082	.604	-.120	.397	-.226	.292	-.170	.295	-.262
-.030	-.091	.737	.136	.785	.164	.501	-.195	.403	-.199	.396	-.303
.128	-.208			.967	.189	.603	.012	.489	-.088	.497	-.251
.418	-.140			1.000	-.069	.703	.060	.594	-.218	.597	-.135
.564	-.019					.784	.121	.700	.028	.702	.030
.710	.117					.868	.208	.786	.161	.786	.117
.976	.293					.923	.247	.858	.234	.864	.162
1.072	.263							.919	.253	.912	.174
1.110	.186							.967	.128		
0.000	0.000										
CN=	.4149		.4136		.4186		.4016		.3700		.3225
CM=	-.0286		-.0655		-.0828		-.0821		-.0986		-.0824

TABLE V — Continued

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(i) M = 1.00. Continued.

$\alpha = 3.44^o$; $C_L = 0.423$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.053	-.021	-.628	.023	-.625	.025	-.595	.022	-.461	.018	-.377
-.567	-.182	.035	-.753	.068	-.759	.079	-.700	.075	-.649	.077	-.600
-.452	-.231	.105	-.681	.134	-.745	.133	-.698	.129	-.645	.129	-.622
-.311	-.300	.178	-.485	.209	-.721	.214	-.701	.201	-.667	.209	-.608
-.023	-.381	.286	-.425	.294	-.650	.295	-.663	.294	-.612	.293	-.580
.133	-.300	.396	-.413	.404	-.531	.407	-.639	.397	-.617	.494	-.567
.272	-.259	.514	-.294	.497	-.303	.502	-.441	.495	-.566	.590	-.567
.416	-.239	.618	-.277	.599	-.293	.601	-.310	.594	-.516	.693	-.587
.565	-.126	.733	-.286	.700	-.422	.698	-.392	.693	-.423	.777	-.466
.713	-.069	.835	-.306	.864	-.420	.863	-.294	.784	-.473	.861	-.140
.854	-.135	.919	-.180	.926	-.130	.923	-.129	.856	-.295	.918	-.098
.980	-.180	.987	-.040	.975	-.098	.977	-.104	.926	-.140	.972	-.025
1.074	-.248							.977	-.094		
1.122	-.147										
				LOWER SURFACE							
-.660	.087	-.022	.305	.024	.220	.025	.106	.019	.015	.020	.074
-.616	.057	.038	.065	.075	.006	0.000	0.000	.066	-.214	.076	-.222
0.000	0.000	.101	-.000	.297	-.095	.130	-.100	.136	-.207	.136	-.216
-.329	-.010	.185	-.048	.400	-.106	.298	-.170	.214	-.129	.221	-.219
-.172	.025	.398	-.052	.604	-.081	.397	-.229	.292	-.120	.295	-.254
-.030	-.081	.737	.141	.785	.165	.501	-.177	.403	-.199	.396	-.287
.128	-.197			.967	.189	.603	.028	.489	-.094	.497	-.250
.418	-.088			1.000	-.083	.703	.060	.594	-.209	.597	-.144
.564	.001					.784	.118	.700	.037	.702	.023
.710	.132					.868	.210	.786	.159	.786	.100
.976	.301					.923	.259	.858	.232	.864	.154
1.072	.268							.919	.253	.912	.165
1.110	.189							.967	.127		
0.000	0.000										
CN=	.4467		.4565		.4886		.4492		.4477		.3808
CM=	-.0325		-.0660		-.0895		-.0834		-.1075		-.0926

$\alpha = 3.94^o$; $C_L = 0.486$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
				UPPER SURFACE							
-.660	-.056	-.021	-.701	.023	-.682	.025	-.637	.022	-.507	.018	-.426
-.567	-.201	.035	-.793	.068	-.812	.079	-.742	.075	-.692	.077	-.648
-.452	-.251	.105	-.771	.134	-.793	.133	-.749	.129	-.693	.129	-.665
-.311	-.318	.178	-.548	.209	-.770	.214	-.742	.201	-.702	.209	-.648
-.023	-.392	.286	-.442	.294	-.724	.295	-.694	.294	-.663	.293	-.618
.133	-.308	.396	-.439	.404	-.667	.407	-.683	.397	-.669	.494	-.619
.272	-.271	.514	-.316	.497	-.357	.502	-.679	.495	-.631	.590	-.598
.416	-.249	.618	-.301	.599	-.314	.601	-.540	.594	-.624	.693	-.624
.565	-.124	.733	-.297	.700	-.435	.698	-.397	.693	-.657	.777	-.268
.713	-.078	.835	-.325	.864	-.409	.863	-.245	.784	-.478	.861	-.177
.854	-.142	.919	-.182	.926	-.132	.923	-.131	.856	-.244	.918	-.141
.980	-.194	.987	-.054	.975	-.101	.977	-.108	.926	-.208	.972	-.129
1.074	-.256							.977	-.190		
1.122	-.166										
				LOWER SURFACE							
-.660	.107	-.022	.353	.024	.274	.025	.154	.019	.080	.020	.115
-.616	.081	.038	.145	.075	.056	0.000	0.000	.066	-.113	.076	-.118
0.000	0.000	.101	.052	.297	-.077	.130	-.056	.136	-.128	.136	-.184
-.329	.005	.185	-.001	.400	-.075	.298	-.158	.214	-.108	.221	-.202
-.172	.042	.398	-.031	.604	-.028	.397	-.212	.292	-.140	.295	-.248
-.030	-.068	.737	.149	.785	.167	.501	-.158	.403	-.199	.396	-.221
.128	-.186			.967	.190	.603	.037	.489	-.083	.497	-.258
.418	-.044			1.000	-.086	.703	.058	.594	-.196	.597	-.166
.564	.023					.784	.117	.700	.039	.702	-.007
.710	.150					.868	.214	.786	.154	.786	.080
.976	.314					.923	.270	.858	.228	.864	.125
1.072	.273							.919	.237	.912	.129
1.110	.196							.967	.086		
0.000	0.000										
CN=	.4976		.5179		.5584		.5253		.5261		.4113
CM=	-.0314		-.0673		-.0956		-.0937		-.1221		-.0867

[redacted]

TABLE V – Concluded

PRESSURE DISTRIBUTIONS OVER WING WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h \doteq -2.5^{\circ}$

(i) M = 1.00. Concluded.

$\alpha = 4.96^{\circ}$; $C_L = 0.598$

STA	.133	STA	.307	STA	.480	STA	.653	STA	.804	STA	.933
X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP	X/C	CP
					UPPER SURFACE						
-.660	-.059	-.021	-.805	.023	-.777	.025	-.736	.022	-.631	.018	-.563
-.567	-.212	.035	-.944	.068	-.898	.079	-.814	.075	-.773	.077	-.729
-.452	-.259	.105	-.892	.134	-.876	.133	-.821	.129	-.770	.129	-.757
-.311	-.314	.178	-.681	.209	-.850	.214	-.813	.201	-.783	.209	-.733
-.023	-.390	.286	-.471	.294	-.817	.295	-.779	.294	-.737	.293	-.702
.133	-.322	.396	-.490	.404	-.803	.407	-.775	.397	-.738	.494	-.701
.272	-.282	.514	-.388	.497	-.699	.502	-.760	.495	-.724	.590	-.684
.416	-.244	.618	-.347	.599	-.382	.601	-.747	.594	-.718	.693	-.696
.565	-.152	.733	-.339	.700	-.455	.698	-.769	.693	-.513	.777	-.717
.713	-.111	.835	-.341	.864	-.304	.863	-.340	.784	-.300	.861	-.289
.854	-.170	.919	-.179	.926	-.133	.923	-.299	.856	-.280	.918	-.261
.980	-.223	.987	-.072	.975	-.111	.977	-.300	.926	-.283	.972	-.239
1.074	-.293							.977	-.282		
1.122	-.197										
					LOWER SURFACE						
-.660	.128	-.022	.412	.024	.351	.025	.265	.019	.183	.020	.228
-.616	.119	.038	.206	.075	.119	0.000	0.000	.066	-.028	.076	-.054
C.000	0.000	.101	.114	.297	-.033	.130	-.003	.136	-.064	.136	-.152
-.329	.038	.185	.059	.400	-.055	.298	-.116	.214	-.083	.221	-.180
-.172	.066	.398	.011	.604	-.034	.397	-.039	.292	-.116	.295	-.221
-.030	-.044	.737	.159	.785	.181	.501	-.160	.403	-.169	.396	-.273
.128	-.149			.967	.189	.603	.044	.489	-.049	.497	-.257
.418	-.004			1.000	-.091	.703	.053	.594	-.198	.597	-.201
.564	.053					.784	.105	.700	-.013	.702	-.043
.710	.173					.868	.218	.786	.118	.786	.043
.976	.326					.923	.275	.858	.199	.864	.087
1.072	.280							.919	.211	.912	.094
1.110	.193							.967	.048		
0.000	0.000										
CN=	.5641		.6215		.6577		.7081		.5752		.5271
CM=	-.0396		-.0713		-.0995		-.1438		-.1111		-.1150

[redacted]

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TABLE VI

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(a) M = 0.25

Fuselage station		C_p for –											
cm	in.	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$
		$\alpha = -5.17^o$; $C_L = -0.432$				$\alpha = 2.48^o$; $C_L = 0.267$				$\alpha = 8.93^o$; $C_L = 0.800$			
153.0	60.22	0.064	0.057	-0.019	0.022	0.040	0.027	0.007	0.041	0.012	-0.013	0.044	0.078
154.4	60.80	.084	.095	.007	.040	.060	.070	.030	.056	.045	.027	.047	.087
156.1	61.46	.119	.095	.032	.053	.094	.072	.053	.065	.080	.046	.068	.093
157.6	62.04	.139	.105	.054	.068	.120	.085	.069	.077	.107	.066	.084	.101
159.2	62.67	.109	.082	.062	.049	.101	.068	.070	.054	.085	.051	.079	.065
160.8	63.30	.059			.009	.053			.024	.038			.024
		$\alpha = -4.17^o$; $C_L = -0.342$				$\alpha = 2.88^o$; $C_L = 0.300$				$\alpha = 9.59^o$; $C_L = 0.851$			
153.0	60.22	0.060	0.047	-0.020	0.026	0.033	0.024	0.015	0.045	0.006	-0.012	0.041	0.078
154.4	60.80	.081	.089	.007	.043	.063	.066	.029	.063	.034	.043	.056	.092
156.1	61.46	.114	.094	.036	.056	.099	.069	.052	.067	.074	.050	.067	.092
157.6	62.04	.135	.103	.056	.074	.119	.087	.066	.082	.099	.071	.084	.100
159.2	62.67	.107	.081	.063	.046	.096	.071	.075	.054	.078	.049	.083	.066
160.8	63.30	.053			.014	.048			.025	.039			.016
		$\alpha = -3.18^o$; $C_L = -0.250$				$\alpha = 3.35^o$; $C_L = 0.341$				$\alpha = 10.49^o$; $C_L = 0.921$			
153.0	60.22	0.058	0.052	-0.014	0.028	0.036	0.019	0.017	0.053	0.004	-0.022	0.052	0.084
154.4	60.80	.083	.086	.009	.048	.059	.070	.032	.065	.031	.032	.059	.096
156.1	61.46	.109	.091	.037	.058	.090	.072	.053	.071	.070	.041	.074	.097
157.6	62.04	.131	.100	.053	.072	.115	.089	.070	.081	.103	.061	.082	.105
159.2	62.67	.109	.072	.065	.048	.098	.072	.076	.055	.080	.046	.085	.067
160.8	63.30	.054			.005	.048			.025	.034			.019
		$\alpha = -2.21^o$; $C_L = -0.161$				$\alpha = 3.88^o$; $C_L = 0.388$				$\alpha = 11.51^o$; $C_L = 0.996$			
153.0	60.22	0.055	0.046	-0.007	0.031	0.029	0.017	0.014	0.047	-0.003	-0.028	0.057	0.087
154.4	60.80	.076	.081	.008	.046	.058	.061	.030	.059	.027	.028	.064	.100
156.1	61.46	.104	.086	.036	.061	.088	.067	.056	.069	.068	.034	.075	.098
157.6	62.04	.125	.095	.052	.075	.116	.080	.068	.079	.100	.054	.086	.110
159.2	62.67	.106	.078	.066	.050	.091	.063	.073	.055	.077	.042	.085	.068
160.8	63.30	.058			.025	.046			.024	.031			.021
		$\alpha = -1.18^o$; $C_L = -0.064$				$\alpha = 4.89^o$; $C_L = 0.474$				$\alpha = 12.43^o$; $C_L = 1.062$			
153.0	60.22	0.051	0.033	-0.008	0.029	0.029	0.014	0.022	0.055	-0.008	-0.037	0.056	0.090
154.4	60.80	.076	.078	.013	.043	.055	.054	.040	.071	.024	.028	.064	.106
156.1	61.46	.103	.080	.043	.053	.092	.068	.060	.073	.068	.034	.076	.100
157.6	62.04	.125	.094	.055	.070	.112	.084	.071	.087	.092	.051	.091	.112
159.2	62.67	.102	.077	.069	.050	.094	.061	.077	.058	.073	.042	.087	.070
160.8	63.30	.053			.025	.045			.023	.031			.023
		$\alpha = -0.05^o$; $C_L = -0.041$				$\alpha = 5.96^o$; $C_L = 0.564$				$\alpha = 13.34^o$; $C_L = 1.129$			
153.0	60.22	0.050	0.037	-0.002	0.037	0.023	0.008	0.025	0.060	-0.019	-0.047	0.053	0.080
154.4	60.80	.073	.079	.019	.050	.058	.057	.038	.072	.007	.019	.070	.105
156.1	61.46	.105	.076	.045	.059	.086	.062	.057	.083	.048	.021	.077	.096
157.6	62.04	.126	.091	.065	.071	.111	.078	.072	.092	.079	.042	.083	.108
159.2	62.67	.101	.073	.069	.053	.090	.066	.077	.058	.066	.038	.086	.068
160.8	63.30	.052			.026	.045			.023	.010			.010
		$\alpha = 0.93^o$; $C_L = 0.127$				$\alpha = 7.01^o$; $C_L = 0.650$				$\alpha = 14.13^o$; $C_L = 1.171$			
153.0	60.22	0.043	0.034	0.004	0.041	0.019	-0.001	0.028	0.067	-0.024	-0.037	0.047	0.110
154.4	60.80	.068	.073	.025	.056	.045	.054	.048	.079	.017	.011	.071	.099
156.1	61.46	.102	.079	.047	.062	.081	.060	.063	.079	.052	.018	.069	.108
157.6	62.04	.126	.091	.063	.076	.110	.074	.077	.095	.094	.029	.087	.107
159.2	62.67	.099	.075	.068	.056	.086	.062	.081	.059	.071	.040	.068	.068
160.8	63.30	.053			.029	.044			.014	.010			.003
		$\alpha = 2.03^o$; $C_L = 0.223$				$\alpha = 7.99^o$; $C_L = 0.728$							
153.0	60.22	0.041	0.027	0.011	0.046	0.014	-0.008	0.036	0.069				
154.4	60.80	.066	.068	.030	.055	.045	.040	.046	.080				
156.1	61.46	.097	.072	.053	.063	.078	.056	.064	.088				
157.6	62.04	.119	.086	.068	.076	.102	.072	.081	.098				
159.2	62.67	.101	.067	.077	.054	.086	.058	.081	.063				
160.8	63.30	.052			.027	.042			.018				

[redacted]

TABLE VI – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50

Fuselage station		C_p for –											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -5.24^{\circ}$; $C_L = -0.459$				$\alpha = 1.98^{\circ}$; $C_L = 0.206$				$\alpha = 6.91^{\circ}$; $C_L = 0.635$			
153.0	60.22	0.064	0.058	-0.029	0.018	0.038	0.025	0.004	0.042	0.017	-0.001	0.025	0.064
154.4	60.80		.080	-.000	.039		.057	.026	.056		.032	.041	.076
156.1	61.46	.120	.096	.031	.053	.101	.078	.052	.064	.088	.058	.063	.080
157.6	62.04	.145	.107	.053	.074	.124	.096	.070	.079	.110	.078	.078	.096
159.2	62.67	.113	.083	.064	.045	.103	.078	.077	.056	.093	.063	.081	.060
160.8	63.30	.054			.010	.054			.024	.044			.017
		$\alpha = -4.18^{\circ}$; $C_L = -0.359$				$\alpha = 2.48^{\circ}$; $C_L = 0.252$				$\alpha = 7.99^{\circ}$; $C_L = 0.723$			
153.0	60.22	0.058	0.052	-0.025	0.021	0.036	0.022	0.009	0.040	0.014	-0.004	0.035	0.066
154.4	60.80		.077	.004	.042		.052	.027	.057		.027	.047	.083
156.1	61.46	.116	.095	.037	.053	.100	.074	.054	.070	.086	.058	.070	.086
157.6	62.04	.138	.104	.056	.073	.123	.092	.071	.079	.112	.080	.083	.101
159.2	62.67	.112	.083	.066	.045	.103	.074	.078	.055	.091	.067	.081	.065
160.8	63.30	.057			.011	.054			.028	.043			.025
		$\alpha = -3.18^{\circ}$; $C_L = -0.267$				$\alpha = 2.90^{\circ}$; $C_L = 0.291$				$\alpha = 8.94^{\circ}$; $C_L = 0.801$			
153.0	60.22	0.055	0.046	-0.018	0.021	0.035	0.022	0.011	0.043	0.011	-0.011	0.043	0.071
154.4	60.80		.070	.002	.045		.046	.030	.059		.021	.051	.085
156.1	61.46	.110	.092	.035	.055	.100	.075	.056	.067	.081	.050	.068	.092
157.6	62.04	.133	.100	.053	.074	.123	.089	.070	.078	.111	.072	.082	.104
159.2	62.67	.111	.082	.065	.048	.104	.075	.079	.058	.089	.058	.084	.068
160.8	63.30	.052			.016	.054			.026	.037			.013
		$\alpha = -2.14^{\circ}$; $C_L = -0.170$				$\alpha = 3.38^{\circ}$; $C_L = 0.337$				$\alpha = 9.93^{\circ}$; $C_L = 0.875$			
153.0	60.22	0.054	0.045	-0.010	0.030	0.033	0.016	0.010	0.047	-0.003	-0.026	0.040	0.075
154.4	60.80		.070	.013	.047		.046	.031	.060		.014	.052	.085
156.1	61.46	.113	.091	.040	.058	.099	.073	.053	.070	.072	.044	.069	.090
157.6	62.04	.140	.103	.060	.075	.122	.090	.070	.081	.106	.067	.084	.103
159.2	62.67	.111	.083	.073	.058	.096	.073	.075	.052	.079	.048	.083	.063
160.8	63.30	.058			.024	.053			.026	.030			.012
		$\alpha = 1.19^{\circ}$; $C_L = -0.082$				$\alpha = 3.94^{\circ}$; $C_L = 0.386$				$\alpha = 10.92^{\circ}$; $C_L = 0.942$			
153.0	60.22	0.047	0.036	-0.015	0.029	0.030	0.018	0.011	0.050	0.007	-0.027	0.050	0.091
154.4	60.80		.064	.017	.046		.047	.034	.067		.020	.055	.103
156.1	61.46	.104	.083	.041	.056	.093	.073	.057	.068	.072	.047	.077	.107
157.6	62.04	.129	.099	.062	.074	.121	.089	.070	.083	.108	.070	.088	.109
159.2	62.67	.107	.078	.069	.051	.098	.073	.078	.054	.083	.053	.091	.075
160.8	63.30	.052			.027	.050			.025	.039			.017
		$\alpha = 0^{\circ}$; $C_L = 0.029$				$\alpha = 4.96^{\circ}$; $C_L = 0.474$							
153.0	60.22	0.046	0.035	-0.004	0.036	0.027	0.016	0.020	0.056				
154.4	60.80		.060	.018	.051		.041	.040	.070				
156.1	61.46	.107	.080	.047	.063	.092	.067	.059	.074				
157.6	62.04	.132	.099	.066	.072	.119	.089	.077	.091				
159.2	62.67	.106	.079	.074	.052	.095	.068	.077	.060				
160.8	63.30	.054			.025	.049			.024				
		$\alpha = 0.97^{\circ}$; $C_L = 0.117$				$\alpha = 5.99^{\circ}$; $C_L = 0.561$							
153.0	60.22	0.043	0.031	-0.001	0.036	0.020	0.004	0.020	0.057				
154.4	60.80		.053	.023	.054		.037	.040	.068				
156.1	61.46	.105	.080	.048	.061	.087	.061	.059	.075				
157.6	62.04	.126	.092	.066	.075	.112	.081	.078	.091				
159.2	62.67	.105	.075	.075	.054	.092	.061	.078	.056				
160.8	63.30	.054			.029	.046			.018				

[redacted]

TABLE VI – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80

Fuselage station		C_p for –											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -5.07^{\circ}$; $C_L = -0.481$				$\alpha = 1.92^{\circ}$; $C_L = 0.213$				$\alpha = 6.96^{\circ}$; $C_L = 0.701$			
153.0	60.22	0.055	0.054	-0.041	0.013	0.036	0.017	-0.004	0.041	0.005	-0.015	0.023	0.059
154.4	60.80	.099	.106	-.004	.042	.078	.092	.031	.056	.057	.076	.049	.081
156.1	61.46	.138	.110	.043	.054	.114	.094	.059	.073	.100	.078	.071	.082
157.6	62.04	.171	.125	.071	.085	.146	.116	.086	.093	.131	.102	.087	.104
159.2	62.67	.140	.104	.079	.061	.129	.098	.095	.074	.114	.087	.095	.072
160.8	63.30	.071			.025	.074			.043	.062			.025
		$\alpha = -4.10^{\circ}$; $C_L = -0.399$				$\alpha = 2.40^{\circ}$; $C_L = 0.260$				$\alpha = 7.99^{\circ}$; $C_L = 0.779$			
153.0	60.22	0.061	0.054	-0.027	0.016	0.033	0.009	0.001	0.040	0.003	-0.018	0.029	0.068
154.4	60.80	.094	.110	.005	.049	.075	.093	.030	.062	.051	.075	.050	.086
156.1	61.46	.136	.110	.048	.060	.116	.088	.060	.073	.097	.075	.076	.089
157.6	62.04	.167	.126	.073	.089	.144	.116	.086	.095	.132	.099	.092	.113
159.2	62.67	.140	.102	.086	.071	.123	.092	.090	.064	.114	.085	.100	.068
160.8	63.30	.073			.029	.071			.038	.059			.025
		$\alpha = -3.16^{\circ}$; $C_L = -0.307$				$\alpha = 2.86^{\circ}$; $C_L = 0.306$				$\alpha = 8.99^{\circ}$; $C_L = 0.830$			
153.0	60.22	0.057	0.041	-0.034	0.017	0.030	0.013	-0.004	0.037	-0.002	-0.026	0.033	0.075
154.4	60.80	.087	.102	.006	.042	.069	.089	.023	.061	.045	.066	.057	.091
156.1	61.46	.127	.106	.047	.062	.111	.091	.060	.074	.096	.068	.077	.093
157.6	62.04	.162	.118	.071	.084	.142	.118	.087	.096	.125	.101	.094	.111
159.2	62.67	.133	.100	.082	.064	.126	.094	.090	.064	.100	.081	.098	.074
160.8	63.30	.066			.030	.070			.030	.051			.020
		$\alpha = -2.18^{\circ}$; $C_L = -0.208$				$\alpha = 3.34^{\circ}$; $C_L = 0.354$				$\alpha = 9.97^{\circ}$; $C_L = 0.882$			
153.0	60.22	0.051	0.043	-0.017	0.022	0.024	0.010	0.001	0.046	-0.017	-0.032	0.038	0.075
154.4	60.80	.092	.108	.015	.052	.063	.087	.031	.066	.043	.070	.050	.090
156.1	61.46	.130	.110	.054	.068	.112	.087	.065	.078	.081	.067	.078	.110
157.6	62.04	.153	.125	.080	.087	.147	.110	.086	.095	.124	.097	.093	.120
159.2	62.67	.135	.103	.090	.067	.126	.093	.091	.063	.113	.087	.099	.080
160.8	63.30	.074			.038	.070			.034	.058			.025
		$\alpha = -1.17$; $C_L = -0.102$				$\alpha = 3.90^{\circ}$; $C_L = 0.408$				$\alpha = 11.01^{\circ}$; $C_L = 0.949$			
153.0	60.22	0.042	0.036	-0.016	0.021	0.024	0.009	0.007	0.046	-0.013	-0.022	0.047	0.096
154.4	60.80	.082	.104	.017	.048	.067	.089	.041	.069	.035	.071	.062	.109
156.1	61.46	.123	.104	.057	.067	.112	.087	.065	.080	.090	.074	.092	.123
157.6	62.04	.152	.123	.078	.083	.143	.111	.087	.099	.129	.095	.105	.127
159.2	62.67	.129	.102	.086	.067	.125	.094	.095	.075	.110	.086	.104	.088
160.8	63.30				.038	.071			.033	.060			.024
		$\alpha = -0.01^{\circ}$; $C_L = 0.017$				$\alpha = 4.83^{\circ}$; $C_L = 0.502$							
153.0	60.22	0.042	0.035	-0.007	0.029	0.019	0.004	0.012	0.044				
154.4	60.80	.078	.101	.022	.050	.060	.080	.041	.066				
156.1	61.46	.121	.100	.059	.066	.109	.079	.068	.081				
157.6	62.04	.151	.123	.086	.090	.134	.109	.088	.100				
159.2	62.67	.130	.100	.093	.070	.113	.089	.090	.074				
160.8	63.30	.078			.044	.064			.025				
		$\alpha = 0.96^{\circ}$; $C_L = 0.115$				$\alpha = 5.82^{\circ}$; $C_L = 0.599$							
153.0	60.22	0.039	0.019	-0.003	0.033	0.014	-0.005	0.014	0.055				
154.4	60.80	.080	.096	.030	.057	.062	.080	.046	.072				
156.1	61.46	.119	.096	.060	.071	.102	.078	.071	.086				
157.6	62.04	.146	.116	.082	.083	.132	.101	.090	.101				
159.2	62.67	.127	.096	.092	.069	.112	.084	.095	.072				
160.8	63.30	.072			.037	.063			.025				

[redacted]

TABLE VI – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^o$

(d) M = 0.90

Fuselage station		C_p for –											
cm	in.	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$
		$\alpha = -5.00^o$; $C_L = -0.514$				$\alpha = 1.00^o$; $C_L = 0.115$				$\alpha = 4.91^o$; $C_L = 0.562$			
153.0	60.22	0.057	0.036	-0.031	0.016	0.035	0.014	-0.007	0.035	0.011	-0.008	0.005	0.044
154.4	60.80		.086	.011	.050		.069	.035	.059		.050	.040	.070
156.1	61.46	.147	.125	.061	.072	.127	.109	.069	.082	.115	.100	.071	.089
157.6	62.04	.184	.144	.085	.106	.159	.138	.095	.102	.150	.128	.099	.112
159.2	62.67	.152	.123	.102	.081	.145	.118	.108	.088	.135	.114	.108	.084
160.8	63.30	.086			.041	.093			.061	.085			.046
		$\alpha = -3.98^o$; $C_L = -0.428$				$\alpha = 1.99^o$; $C_L = 0.224$				$\alpha = 5.90^o$; $C_L = 0.675$			
153.0	60.22	0.053	0.036	-0.030	0.017	0.036	0.012	-0.001	0.035	0.005	-0.009	0.011	0.051
154.4	60.80		.082	.015	.051		.067	.031	.062		.047	.044	.079
156.1	61.46	.147	.119	.060	.076	.128	.106	.070	.079	.114	.097	.078	.094
157.6	62.04	.177	.143	.089	.104	.161	.136	.097	.103	.147	.127	.100	.113
159.2	62.67	.151	.120	.106	.083	.141	.115	.107	.082	.129	.113	.111	.084
160.8	63.30	.087			.046	.093			.059	.082			.042
		$\alpha = -3.13^o$; $C_L = -0.345$				$\alpha = 2.46^o$; $C_L = 0.275$				$\alpha = 6.94^o$; $C_L = 0.776$			
153.0	60.22	0.053	0.034	-0.028	0.020	0.028	0.009	-0.006	0.033	0.009	-0.021	0.015	0.061
154.4	60.80		.077	.012	.054		.062	.037	.061		.040	.050	.084
156.1	61.46	.141	.117	.059	.075	.122	.102	.072	.083	.112	.094	.081	.095
157.6	62.04	.171	.143	.086	.106	.157	.134	.096	.102	.152	.127	.103	.121
159.2	62.67	.149	.119	.104	.084	.140	.120	.109	.083	.129	.114	.107	.086
160.8	63.30	.086			.046	.090			.056	.081			.041
		$\alpha = -2.12^o$; $C_L = -0.234$				$\alpha = 2.98^o$; $C_L = 0.332$				$\alpha = 7.97^o$; $C_L = 0.834$			
153.0	60.22	0.045	0.029	-0.027	0.014	0.027	0.011	-0.005	0.038	-0.002	-0.015	0.025	0.066
154.4	60.80		.072	.017	.049		.055	.033	.063		.038	.049	.088
156.1	61.46	.139	.115	.062	.077	.122	.107	.071	.077	.114	.098	.079	.103
157.6	62.04	.169	.136	.088	.100	.157	.134	.097	.100	.146	.131	.107	.127
159.2	62.67	.151	.122	.102	.082	.139	.114	.108	.082	.132	.113	.110	.090
160.8	63.30	.095			.049	.086			.050	.078			.037
		$\alpha = -1.16^o$; $C_L = -0.126$				$\alpha = 3.43^o$; $C_L = 0.383$							
153.0	60.22	0.042	0.025	-0.019	0.021	0.030	0.008	0.001	0.040				
154.4	60.80		.073	.025	.052		.062	.043	.068				
156.1	61.46	.132	.111	.062	.074	.127	.105	.073	.087				
157.6	62.04	.165	.140	.092	.096	.159	.133	.098	.107				
159.2	62.67	.151	.122	.106	.080	.143	.119	.108	.082				
160.8	63.30	.094			.061	.093			.059				
		$\alpha = 0.02^o$; $C_L = 0.008$				$\alpha = 3.93^o$; $C_L = 0.443$							
153.0	60.22	0.035	0.020	-0.018	0.022	0.022	0.007	0.000	0.042				
154.4	60.80		.069	.023	.050		.056	.037	.068				
156.1	61.46	.132	.110	.067	.075	.120	.103	.071	.081				
157.6	62.04	.158	.138	.091	.098	.150	.130	.097	.108				
159.2	62.67	.144	.119	.103	.080	.137	.117	.107	.085				
160.8	63.30	.092			.055	.091			.051				

[redacted]

TABLE VI – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(e) M = 0.95

Fuselage station		C_p for –											
cm	in.	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$
		$\alpha = -4.97^\circ$; $C_L = -0.546$				$\alpha = 1.02^\circ$; $C_L = 0.122$				$\alpha = 4.94^\circ$; $C_L = 0.599$			
153.0	60.22	0.052	0.035	-0.057	-0.007	0.029	0.005	-0.016	0.027	-0.011	-0.021	-0.000	0.049
154.4	60.80		.090	.012	.046		.068	.034	.057		.053	.040	.072
156.1	61.46	.162	.136	.067	.086	.139	.119	.076	.088	.123	.108	.082	.096
157.6	62.04	.203	.158	.100	.117	.168	.149	.105	.110	.159	.144	.105	.116
159.2	62.67	.173	.138	.124	.100	.158	.130	.119	.097	.148	.127	.118	.095
160.8	63.30	.102			.066	.112			.076	.101			.067
		$\alpha = -3.97^\circ$; $C_L = -0.471$				$\alpha = 1.99^\circ$; $C_L = 0.238$				$\alpha = 5.87^\circ$; $C_L = 0.689$			
153.0	60.22	0.044	0.024	-0.051	-0.009	0.014	-0.002	-0.015	0.025	-0.010	-0.024	0.002	0.046
154.4	60.80		.082	.016	.045		.067	.034	.056		.047	.039	.073
156.1	61.46	.153	.128	.067	.078	.130	.115	.077	.084	.120	.106	.078	.091
157.6	62.04	.192	.154	.095	.110	.164	.146	.108	.106	.159	.138	.104	.116
159.2	62.67	.171	.135	.117	.096	.150	.131	.117	.094	.145	.130	.116	.094
160.6	63.30	.097			.060	.106			.075	.102			.062
		$\alpha = -2.99^\circ$; $C_L = -0.368$				$\alpha = 2.48^\circ$; $C_L = 0.295$				$\alpha = 6.92^\circ$; $C_L = 0.789$			
153.0	60.22	0.042	0.025	-0.036	0.005	0.022	0.002	-0.011	0.032	-0.013	-0.027	0.006	0.052
154.4	60.80		.080	.025	.052		.067	.036	.062		.047	.035	.080
156.1	61.46	.150	.127	.073	.085	.134	.116	.081	.089	.116	.099	.073	.095
157.6	62.04	.183	.155	.101	.117	.166	.148	.108	.109	.155	.135	.101	.124
159.2	62.67	.167	.135	.117	.101	.154	.133	.123	.098	.149	.129	.113	.095
160.8	63.30	.109			.072	.108			.075	.100			.055
		$\alpha = -2.12^\circ$; $C_L = -0.270$				$\alpha = 2.84^\circ$; $C_L = 0.341$				$\alpha = 8.01^\circ$; $C_L = 0.876$			
153.0	60.22	0.035	0.018	-0.030	0.007	0.022	0.005	-0.008	0.026	-0.022	-0.029	0.005	0.058
154.4	60.80		.076	.023	.050		.067	.039	.061		.038	.036	.085
156.1	61.46	.140	.122	.074	.083	.131	.118	.080	.086	.114	.096	.070	.096
157.6	62.04	.176	.150	.101	.109	.166	.152	.107	.110	.151	.136	.101	.118
159.2	62.67	.160	.133	.117	.097	.157	.136	.121	.096	.147	.121	.109	.098
160.8	63.30	.106			.073	.107			.071	.095			.054
		$\alpha = -1.19^\circ$; $C_L = -0.149$				$\alpha = 3.42^\circ$; $C_L = 0.412$				$\alpha = 8.55^\circ$; $C_L = 0.917$			
153.0	60.22	0.033	0.012	-0.025	0.014	0.011	-0.000	-0.004	0.038	-0.035	-0.037	0.015	0.063
154.4	60.80		.075	.028	.054		.068	.038	.068		.039	.045	.089
156.1	60.46	.138	.121	.074	.085	.133	.122	.080	.094	.109	.095	.074	.097
157.6	62.04	.173	.150	.104	.107	.171	.150	.108	.112	.157	.130	.098	.132
159.2	62.67	.163	.131	.114	.100	.157	.134	.122	.097	.145	.126	.121	.095
160.8	63.30	.110		.076	.076	.107			.072	.099			.052
		$\alpha = -0.03^\circ$; $C_L = -0.003$				$\alpha = 3.94^\circ$; $C_L = 0.478$							
153.0	60.22	0.031	0.009	-0.018	0.019	0.010	-0.008	-0.004	0.034				
154.4	60.80		.067	.031	.054		.061	.040	.069				
156.1	60.46	.134	.121	.073	.080	.129	.111	.079	.088				
157.6	62.04	.169	.149	.101	.106	.161	.146	.111	.113				
159.2	62.67	.159	.132	.118	.098	.152	.134	.120	.096				
160.8	63.30	.109			.079	.104			.071				

[redacted]

TABLE VI – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(f) M = 0.97

Fuselage station		C_p for –											
cm	in.	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$
		$\alpha = -4.95^\circ$; $C_L = -0.560$				$\alpha = 1.93^\circ$; $C_L = 0.232$				$\alpha = 6.91^\circ$; $C_L = 0.792$			
153.0	60.22	0.046	0.026	-0.124	-0.080	-0.038	-0.085	-0.036	0.017	-0.051	-0.074	-0.003	0.046
154.4	60.80		.087	-.027	.005		.056	.029	.059		.033	.032	.076
156.1	61.46	.171	.137	.046	.065	.124	.120	.085	.093	.113	.097	.070	.095
157.6	62.04	.226	.166	.092	.111	.165	.156	.114	.117	.154	.138	.098	.127
159.2	62.67	.189	.145	.123	.107	.163	.144	.132	.110	.151	.135	.118	.099
160.8	63.30	.107			.082	.120			.093	.110			.066
		$\alpha = -4.02^\circ$; $C_L = -0.480$				$\alpha = 2.47^\circ$; $C_L = 0.303$				$\alpha = 7.98^\circ$; $C_L = 0.878$			
153.0	60.22	0.033	0.019	-0.118	-0.068	-0.031	-0.068	-0.029	0.013	-0.057	-0.079	-0.001	0.052
154.4	60.80		.079	-.017	.009		.048	.034	.057		.028	.032	.077
156.1	61.46	.159	.126	.051	.065	.127	.122	.084	.089	.107	.090	.068	.097
157.6	62.04	.205	.158	.090	.113	.164	.155	.112	.115	.154	.133	.098	.120
159.2	62.67	.180	.141	.117	.101	.166	.141	.129	.104	.150	.129	.111	.094
160.8	63.30	.104			.072	.117			.089	.106			.056
		$\alpha = -3.03^\circ$; $C_L = -0.380$				$\alpha = 2.88^\circ$; $C_L = 0.358$				$\alpha = 9.08^\circ$; $C_L = 0.954$			
153.0	60.22	0.028	0.017	-0.092	-0.049	-0.040	-0.092	-0.036	0.018	-0.067	-0.092	0.010	0.061
154.4	60.80		.082	-.006	.026		.047	.024	.058		.024	.037	.086
156.1	61.46	.157	.134	.065	.073	.119	.117	.075	.087	.109	.085	.067	.103
157.6	62.04	.197	.160	.102	.117	.159	.151	.110	.114	.152	.136	.101	.127
159.2	62.67	.181	.141	.124	.110	.156	.141	.125	.102	.149	.126	.107	.097
160.8	63.30	.117			.083	.114			.084	.112			.061
		$\alpha = -2.09^\circ$; $C_L = -0.276$				$\alpha = 3.47^\circ$; $C_L = 0.432$				$\alpha = 10.04^\circ$; $C_L = 1.016$			
153.0	60.22	0.005	-0.011	-0.078	-0.037	-0.027	-0.052	-0.025	0.022	-0.082	-0.098	0.020	0.067
154.4	60.80		.076	.003	.030		.054	.030	.063		.026	.042	.088
156.1	61.46	.145	.126	.068	.073	.125	.118	.077	.092	.098	.085	.068	.104
157.6	62.04	.181	.159	.106	.116	.162	.155	.113	.117	.146	.128	.100	.134
159.2	62.67	.174	.144	.124	.109	.160	.144	.130	.104	.147	.125	.108	.099
160.8	63.30	.114			.083	.118			.085	.112			.059
		$\alpha = -1.20^\circ$; $C_L = -0.165$				$\alpha = 3.89^\circ$; $C_L = 0.487$				$\alpha = 11.66^\circ$; $C_L = 1.106$			
153.0	60.22	-0.001	-0.031	-0.070	-0.020	-0.039	-0.081	-0.021	0.029	-0.103	-0.122	0.031	0.083
154.4	60.80		.068	.012	.039		.047	.030	.064		.012	.052	.105
156.1	61.46	.136	.124	.074	.081	.121	.110	.077	.094	.091	.077	.067	.112
157.6	62.04	.175	.160	.111	.113	.161	.151	.113	.119	.143	.122	.095	.135
159.2	62.67	.172	.145	.128	.110	.161	.142	.129	.103	.149	.131	.109	.106
160.8	63.30	.125			.091	.116			.080	.112			.061
		$\alpha = 0^\circ$; $C_L = -0.009$				$\alpha = 4.91^\circ$; $C_L = 0.499$							
153.0	60.22	-0.017	-0.049	-0.056	-0.003	-0.054	-0.102	-0.007	0.037				
154.4	60.80		.063	.019	.042		.037	.039	.073				
156.1	61.46	.134	.124	.072	.083	.114	.109	.082	.100				
157.6	62.04	.170	.160	.108	.113	.159	.153	.116	.122				
159.2	62.67	.173	.144	.127	.106	.156	.143	.128	.107				
160.8	63.30	.125			.090	.115			.077				
		$\alpha = 0.96^\circ$; $C_L = 0.110$				$\alpha = 5.91^\circ$; $C_L = 0.698$							
153.0	60.22	-0.028	-0.068	-0.045	0.004	-0.055	-0.089	-0.011	0.041				
154.4	60.80		.054	.024	.054		.035	.033	.071				
156.1	61.46	.128	.121	.078	.089	.113	.101	.073	.096				
157.6	62.04	.168	.154	.111	.115	.154	.141	.103	.122				
159.2	62.67	.166	.139	.129	.108	.151	.135	.118	.099				
160.8	63.30	.123			.093	.105			.069				

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TABLE VI – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(g) M = 0.98

Fuselage station		C_p for –											
cm	in.	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$
		$\alpha = -1.06^\circ$; $C_L = -0.148$				$\alpha = 1.96^\circ$; $C_L = 0.245$				$\alpha = 3.46^\circ$; $C_L = 0.444$			
153.0	60.22	-0.028	-0.091	-0.095	-0.046	-0.066	-0.139	-0.087	-0.026	-0.071	-0.143	-0.071	-0.013
154.4	60.80		.046	-.011	.019		.020	-.009	.028		.024	-.002	.037
156.1	61.46	.129	.121	.065	.068	.113	.103	.058	.068	.107	.104	.057	.078
157.6	62.04	.172	.154	.102	.105	.153	.146	.102	.105	.153	.145	.100	.111
159.2	62.67		.143	.128	.105		.140	.127	.099		.141	.122	.101
160.8	63.30	.121			.097	.118			.085	.116			.075
		$\alpha = -0.08^\circ$; $C_L = -0.235$				$\alpha = 2.47^\circ$; $C_L = 0.313$				$\alpha = 3.97^\circ$; $C_L = 0.501$			
153.0	60.22	-0.049	-0.120	-0.087	-0.034	-0.060	-0.139	-0.074	-0.021	-0.075	-0.131	-0.058	.004
154.4	60.80		.037	-.016	.021		.027	-.002	.035		.030	.015	.051
156.1	61.46	.119	.114	.060	.069	.111	.107	.064	.073	.107	.102	.069	.085
157.6	62.04	.159	.151	.103	.103	.159	.147	.105	.107	.152	.144	.107	.118
159.2	62.67		.136	.126	.104		.141	.129	.106		.144	.123	.104
160.8	63.30	.120			.090	.122			.089	.117			.082
		$\alpha = 0.96^\circ$; $C_L = 0.114$				$\alpha = 2.94^\circ$; $C_L = 0.378$				$\alpha = 4.95^\circ$ $C_L = 0.601$			
153.0	60.22	-0.054	-0.132	-0.081	-0.028	-0.061	-0.142	-0.072	-0.017	-0.077	-0.137	-0.027	0.026
154.4	60.80		.032	-.001	.030		.031	.005	.039		.026	.026	.066
156.1	61.46	.118	.108	.062	.075	.113	.110	.070	.081	.102	.107	.080	.096
157.6	62.04	.158	.151	.101	.107	.158	.152	.107	.113	.152	.142	.108	.123
159.2	62.67		.141	.125	.103		.143	.127	.106		.141	.126	.107
160.8	63.30	.120			.093	.120			.086	.113			.081

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TABLE VI – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99

Fuselage station		C_p for –											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -4.96^{\circ}$; $C_L = -0.572$				$\alpha = 1.95^{\circ}$; $C_L = 0.226$				$\alpha = 6.82^{\circ}$; $C_L = 0.782$			
153.0	60.22	-0.071	-0.160	-0.138	-0.082	-0.051	-0.134	-0.143	-0.091	-0.055	-0.081	-0.006	0.046
154.4	60.80		.002	-.050	-.025		.021	-.058	-.030		.035	.035	.080
156.1	61.46	.116	.101	.036	.038	.132	.104	.033	.034	.114	.099	.076	.106
157.6	62.04	.171	.145	.088	.089	.182	.144	.082	.083	.159	.144	.106	.131
159.2	62.67	.167	.138	.118	.098	.173	.139	.114	.094	.164	.142	.125	.114
160.8	63.30	.119			.090	.113			.084	.124			.079
		$\alpha = -3.92^{\circ}$; $C_L = -0.488$				$\alpha = 2.50^{\circ}$; $C_L = 0.303$				$\alpha = 7.92^{\circ}$; $C_L = 0.875$			
153.0	60.22	-0.087	-0.178	-0.124	-0.070	-0.102	-0.195	-0.125	-0.057	-0.059	-0.084	0.008	0.061
154.4	60.80		-.017	-.041	-.015		-.014	-.053	-.014		.038	.041	.091
156.1	61.46	.106	.095	.042	.041	.101	.094	.038	.044	.116	.103	.078	.112
157.6	62.04	.156	.140	.092	.088	.146	.144	.086	.089	.166	.150	.110	.139
159.2	62.67	.162	.135	.121	.101	.155	.138	.120	.096	.172	.148	.130	.117
160.8	63.30	.117			.095	.117			.085	.130			.078
		$\alpha = -2.97^{\circ}$; $C_L = -0.392$				$\alpha = 2.91^{\circ}$; $C_L = 0.360$				$\alpha = 9.08^{\circ}$; $C_L = 0.960$			
153.0	60.22	-0.091	-0.188	-0.123	-0.061	-0.102	-0.188	-0.125	-0.057	-0.081	-0.108	0.007	0.061
154.4	60.80		-.022	-.045	-.011		-.006	-.049	-.007		.014	.040	.086
156.1	61.46	.102	.093	.039	.043	.099	.092	.040	.046	.102	.095	.075	.109
157.6	62.04	.150	.139	.090	.091	.149	.141	.086	.094	.153	.134	.103	.134
159.2	62.67	.155	.131	.116	.094	.151	.140	.123	.095	.158	.137	.116	.113
160.8	63.30	.121			.089	.114			.088	.120			.066
		$\alpha = -2.06^{\circ}$; $C_L = -0.294$				$\alpha = 3.46^{\circ}$; $C_L = 0.430$				$\alpha = 10.04^{\circ}$; $C_L = 1.032$			
153.0	60.22	-0.098	-0.191	-0.117	-0.055	-0.098	-0.196	-0.114	-0.045	-0.103	-0.136	0.014	0.065
154.4	60.80		-.016	-.042	-.003		-.006	-.044	.006		.016	.042	.092
156.1	61.46	.099	.089	.041	.055	.096	.092	.040	.053	.094	.083	.069	.103
157.6	62.04	.149	.142	.093	.093	.147	.139	.094	.100	.152	.131	.096	.134
159.2	62.67	.152	.137	.123	.097	.153	.140	.120	.099	.160	.138	.115	.113
160.8	63.30	.117			.090	.118			.085	.126			.061
		$\alpha = -1.14^{\circ}$; $C_L = -0.179$				$\alpha = 3.94^{\circ}$; $C_L = 0.484$				$\alpha = 11.62^{\circ}$; $C_L = 1.117$			
153.0	60.22	0.022	-0.032	-0.155	-0.123	-0.102	-0.203	-0.096	-0.033	-0.123	-0.171	0.022	0.079
154.4	60.80		.071	-.049	-.025		.002	-.029	.013		-.007	.043	.099
156.1	61.46	.181	.135	.031	.042	.097	.089	.054	.063	.095	.070	.070	.112
157.6	62.04	.236	.167	.080	.101	.146	.142	.100	.105	.158	.118	.096	.143
159.2	62.67	.198	.149	.121	.107	.150	.141	.123	.100	.167	.128	.112	.112
160.8	63.30	.119			.086	.118			.085	.125			.063
		$\alpha = 0.02^{\circ}$; $C_L = -0.024$				$\alpha = 4.92^{\circ}$; $C_L = 0.592$							
153.0	60.22	0.004	-0.040	-0.138	-0.104	-0.090	-0.187	-0.058	-0.002				
154.4	60.80		.068	-.053	-.021		.007	.005	.042				
156.1	61.46	.165	.128	.043	.045	.097	.095	.065	.081				
157.6	62.04	.222	.161	.087	.101	.147	.142	.106	.115				
159.2	62.67	.195	.151	.122	.105	.151	.142	.119	.105				
160.8	63.30	.118			.086	.115			.085				
		$\alpha = 1.03^{\circ}$; $C_L = 0.104$				$\alpha = 5.88^{\circ}$; $C_L = 0.688$							
153.0	60.22	-0.023	-0.082	-0.123	-0.079	-0.065	-0.111	-0.026	0.032				
154.4	60.80		.048	-.040	-.015		.029	.024	.068				
156.1	61.46	.147	.123	.044	.054	.109	.101	.073	.099				
157.6	62.04	.198	.160	.088	.096	.157	.145	.107	.126				
159.2	62.67	.183	.145	.119	.102	.156	.143	.129	.107				
160.8	63.30	.115			.084	.121			.079				

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TABLE VI – Concluded

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(i) M = 1.00

Fuselage station		C_p for –											
cm	in.	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$
		$\alpha = -1.06^\circ$; $C_L = -0.161$				$\alpha = 1.97^\circ$; $C_L = 0.219$				$\alpha = 3.47^\circ$; $C_L = 0.418$			
153.0	60.22	-0.091	-0.166	-0.151	-0.090	-0.134	-0.208	-0.132	-0.063	-0.123	-0.207	-0.106	-0.039
154.4	60.80		-.053	-.083	-.042		-.079	-.077	-.027		-.046	-.051	-.006
156.1	61.46	.101	.079	.007	.009	.074	.065	.008	.015	.077	.070	.024	.036
157.6	62.04	.157	.128	.065	.064	.131	.123	.063	.066	.133	.128	.077	.081
159.2	62.67		.123	.101	.075		.123	.103	.074		.127	.112	.085
160.8	63.30	.105			.082	.105			.076	.105			.074
		$\alpha = -0.07^\circ$; $C_L = -0.367$				$\alpha = 2.47^\circ$; $C_L = 0.287$				$\alpha = 3.95^\circ$; $C_L = 0.476$			
153.0	60.22	-0.104	-0.183	-0.144	-0.086	-0.131	-0.216	-0.126	-0.064	-0.110	-0.203	-0.083	-0.021
154.4	60.80		.060	-.080	-.036		-.073	-.073	-.028		-.030	-.021	.017
156.1	61.46	.087	.077	.009	.012	.074	.064	.005	.018	.085	.076	.044	.057
157.6	62.04	.142	.126	.064	.063	.129	.120	.065	.068	.136	.131	.093	.099
159.2	62.67		.124	.100	.074		.121	.099	.078		.132	.116	.095
160.8	63.30	.109			.081	.103			.071	.110			.076
		$\alpha = 0.95^\circ$; $C_L = 0.090$				$\alpha = 2.96^\circ$; $C_L = 0.353$				$\alpha = 4.96^\circ$; $C_L = 0.586$			
153.0	60.22	-0.123	-0.198	-0.142	-0.076	-0.133	-0.209	-0.116	-0.057	-0.092	-0.159	-0.062	0.004
154.4	60.80		-.075	-.079	-.031		-.061	-.064	-.017		.007	.007	.051
156.1	61.46	.079	.066	.003	.011	.071	.069	.015	.025	.103	.093	.069	.090
157.6	62.04	.133	.119	.060	.060	.132	.126	.070	.074	.153	.141	.108	.125
159.2	62.67		.119	.099	.073		.124	.102	.079		.145	.127	.110
160.8	63.30	.102			.078	.108			.080	.120			.085

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TABLE VII

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5.0^o$

(a) M = 0.50

Fuselage station		C_p for –											
cm	in.	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$
		$\alpha = -5.20^o$; $C_L = -0.490$				$\alpha = 1.89^o$; $C_L = 0.163$				$\alpha = 6.90^o$; $C_L = 0.599$			
153.0	60.22	0.084	0.080	-0.055	0.003	0.056	0.047	-0.018	0.025	0.033	0.019	0.000	0.044
154.4	60.80		.099	-.023	.025		.072	.002	.042		.055	.025	.064
156.1	61.46	.140	.115	.015	.035	.117	.094	.032	.050	.101	.076	.042	.070
157.6	62.04	.160	.121	.036	.066	.136	.109	.051	.068	.122	.098	.063	.091
159.2	62.67	.123	.096	.050	.039	.113	.088	.062	.046	.104	.073	.062	.052
160.8	63.30	.056			.001	.058			.018	.047			.010
		$\alpha = -4.16^o$; $C_L = -0.396$				$\alpha = 2.40^o$; $C_L = 0.211$				$\alpha = 7.94^o$; $C_L = 0.680$			
153.0	60.22	0.075	0.075	-0.047	0.008	0.058	0.047	-0.014	0.028	0.031	0.019	0.016	0.051
154.4	60.80		.095	-.015	.029		.070	.010	.042		.050	.024	.069
156.1	61.46	.127	.112	.013	.039	.112	.092	.041	.056	.097	.074	.052	.077
157.6	62.04	.158	.121	.039	.064	.137	.110	.057	.068	.119	.093	.061	.093
159.2	62.67	.126	.093	.054	.041	.116	.088	.065	.045	.101	.069	.067	.057
160.8	63.30	.060			.007	.061			.017	.051			.012
		$\alpha = -3.14^o$; $C_L = -0.302$				$\alpha = 2.88^o$; $C_L = 0.252$				$\alpha = 8.94^o$; $C_L = 0.762$			
153.0	60.22	0.077	0.070	-0.040	0.013	0.055	0.043	-0.012	0.031	0.028	0.008	0.008	0.056
154.4	60.80		.091	-.013	.030		.073	.009	.047		.040	.027	.070
156.1	61.46	.127	.113	.023	.048	.114	.093	.039	.058	.093	.070	.045	.077
157.6	62.04	.151	.119	.047	.067	.134	.103	.058	.067	.119	.093	.058	.094
159.2	62.67	.124	.098	.055	.040	.108	.088	.066	.047	.091	.073	.072	.058
160.8	63.30	.063			.009	.057			.015	.046			.005
		$\alpha = -2.15^o$; $C_L = -0.214$				$\alpha = 3.38^o$; $C_L = 0.296$				$\alpha = 9.92^o$; $C_L = 0.835$			
153.0	60.22	0.074	0.071	-0.033	0.010	0.049	0.038	-0.010	0.030	0.021	0.002	0.023	0.062
154.4	60.80		.088	-.007	.029		.065	.015	.048		.041	.037	.075
156.1	61.46	.126	.104	.022	.045	.111	.090	.039	.055	.084	.072	.055	.082
157.6	62.04	.145	.116	.044	.068	.130	.104	.056	.070	.116	.089	.067	.096
159.2	62.67	.123	.094	.056	.045	.110	.087	.063	.046	090	.070	.072	.061
160.8	63.30	.064			.013	.056			.012	.053			.006
		$\alpha = -1.20^o$; $C_L = -0.124$				$\alpha = 3.95^o$; $C_L = 0.347$				$\alpha = 10.87^o$; $C_L = 0.897$			
153.0	60.22	0.071	0.065	-0.034	0.016	0.048	0.038	-0.010	0.030	0.013	-0.002	0.022	0.070
154.4	60.80		.086	-.003	.031		.066	.016	.045		.041	.036	.090
156.1	61.46	.126	.106	.026	.046	.106	.087	.038	.060	.091	.069	.065	.085
157.6	62.04	.146	.115	.048	.068	.135	.105	.059	.078	.120	.086	.073	.108
159.2	62.67	.121	.092	.061	.048	.113	.078	.066	.048	.100	.074	.074	.070
160.8	63.30	.065			.019	.057			.013	.051			.013
		$\alpha = 0.01^o$; $C_L = -0.007$				$\alpha = 4.94^o$; $C_L = 0.436$							
153.0	60.22	0.061	0.054	-0.026	0.013	0.045	0.030	-0.002	0.038				
154.4	60.80		.080	-.001	.034		.055	.014	.055				
156.1	61.46	.118	.099	.025	.047	.101	.086	.043	.064				
157.6	62.04	.141	.109	.045	.066	.127	.097	.060	.077				
159.2	62.67	.118	.089	.055	.047	.101	.077	.062	.049				
160.8	63.30	.061			.019	.053			.010				
		$\alpha = 1.02^o$; $C_L = 0.085$				$\alpha = 5.99^o$; $C_L = 0.523$							
153.0	60.22	0.063	0.052	-0.023	0.024	0.039	0.029	-0.004	0.044				
154.4	60.80		.077	.005	.040		.055	.018	.056				
156.1	61.46	.117	.098	.033	.047	.099	.079	.043	.066				
157.6	62.04	.139	.109	.054	.066	.126	.094	.065	.085				
159.2	62.67	.114	.091	.064	.046	.101	.076	.068	.051				
160.8	63.30	.065			.020	.050			.010				

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TABLE VII – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5.0^o$

(b) M = 0.80

Fuselage station		C_p for –											
cm	in.	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$
		$\alpha = -4.97^o$; $C_L = -0.514$				$\alpha = 1.88^o$; $C_L = 0.167$				$\alpha = 6.97^o$; $C_L = 0.658$			
153.0	60.22	0.087	0.076	-0.063	-0.007	0.060	0.047	-0.036	0.007	0.031	0.021	-0.007	0.037
154.4	60.80		.104	-.016	.019		.080	.004	.036		.062	.022	.062
156.1	61.46	.157	.128	.021	.048	.130	.109	.042	.057	.115	.089	.043	.070
157.6	62.04	.179	.135	.051	.077	.162	.132	.065	.079	.140	.117	.067	.092
159.2	62.67	.140	.112	.072	.048	.138	.108	.074	.060	.123	.098	.084	.062
160.8	63.30	.066			.015	.080			.033	.064			.019
		$\alpha = -4.09^o$; $C_L = -0.440$				$\alpha = 2.37^o$; $C_L = 0.218$				$\alpha = 8.02^o$; $C_L = 0.733$			
153.0	60.22	0.077	0.072	-0.060	-0.005	0.055	0.049	-0.028	0.016	0.028	0.006	-0.003	0.045
154.4	60.80		.104	-.012	.025		.075	.002	.042		.049	.015	.069
156.1	61.46	.153	.128	.022	.050	.134	.106	.040	.057	.104	.083	.049	.072
157.6	62.04	.177	.138	.052	.084	.160	.126	.068	.080	.143	.109	.071	.096
159.2	62.67	.146	.116	.068	.056	.138	.106	.082	.060	.118	.092	.075	.063
160.8	63.30	.075			.015	.080			.023	.060			.016
		$\alpha = -3.11^o$; $C_L = -0.348$				$\alpha = 2.84^o$; $C_L = 0.262$				$\alpha = 8.97^o$; $C_L = 0.786$			
153.0	60.22	0.077	0.073	-0.056	-0.005	0.055	0.041	-0.025	0.017	0.016	0.005	-0.005	0.042
154.4	60.80		.105	-.016	.024		.079	.009	.047		.044	.014	.063
156.1	61.46	.142	.123	.025	.052	.131	.112	.043	.058	.104	.087	.042	.076
157.6	62.04	.172	.138	.050	.080	.157	.129	.066	.077	.134	.109	.073	.101
159.2	62.67	.144	.114	.063	.058	.139	.108	.080	.062	.108	.086	.070	.060
160.8	63.30	.078			.019	.075			.031	.052			.015
		$\alpha = -2.17^o$; $C_L = -0.251$				$\alpha = 3.35^o$; $C_L = 0.309$				$\alpha = 10.05^o$; $C_L = 0.844$			
153.0	60.22	0.078	0.067	-0.050	0.004	0.056	0.039	-0.021	0.024	0.018	-0.013	0.005	0.050
154.4	60.80		.097	-.008	.031		.075	.014	.043		.042	.022	.076
156.1	61.46	.146	.119	.035	.055	.126	.110	.046	.061	.097	.083	.048	.087
157.6	62.04	.171	.136	.058	.078	.155	.129	.074	.084	.129	.102	.071	.100
159.2	62.67	.143	.115	.073	.058	.138	.107	.081	.057	.109	.080	.081	.063
160.8	63.30	.075			.024	.081			.032	.057			.007
		$\alpha = -1.13^o$; $C_L = -0.144$				$\alpha = 3.99^o$; $C_L = 0.373$				$\alpha = 10.97^o$; $C_L = 0.903$			
153.0	60.22	0.070	0.063	-0.041	0.003	0.045	0.038	-0.025	0.022	0.003	-0.013	0.000	0.064
154.4	60.80		.096	-.006	.035		.073	-.000	.049		.033	.025	.077
156.1	61.46	.142	.122	.036	.048	.124	.107	.041	.062	.098	.074	.059	.088
157.6	62.04	.168	.136	.063	.084	.150	.127	.064	.086	.127	.107	.078	.109
159.2	62.67	.148	.114	.077	.060	.127	.106	.077	.062	.103	.084	.071	.067
160.8	63.30	.079			.030	.073			.034	.052			.000
		$\alpha = 0.04^o$; $C_L = -0.026$				$\alpha = 4.91^o$; $C_L = 0.464$							
153.0	60.22	0.066	0.056	-0.032	0.007	0.046	0.026	-0.006	0.030				
154.4	60.80		.090	-.001	.032		.065	.015	.054				
156.1	61.46	.143	.119	.042	.053	.126	.103	.048	.065				
157.6	62.04	.166	.137	.066	.076	.148	.124	.069	.086				
159.2	62.67	.140	.117	.075	.056	.125	.101	.085	.066				
160.8	63.30	.077			.038	.071			.027				
		$\alpha = 0.99^o$; $C_L = 0.074$				$\alpha = 5.96^o$; $C_L = 0.569$							
153.0	60.22	0.059	0.050	-0.028	0.011	0.040	0.021	-0.007	0.034				
154.4	60.80		.081	-.001	.038		.065	.018	.057				
156.1	61.46	.130	.109	.039	.053	.118	.103	.054	.066				
157.6	62.04	.163	.127	.069	.082	.152	.125	.078	.092				
159.2	62.67	.138	.109	.079	.060	.128	.106	.081	.062				
160.8	63.30	.078			.032	.068			.029				

[redacted]

TABLE VII – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5.0^\circ$

(c) M = 0.90

Fuselage station		C_p for –											
cm	in.	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$
		$\alpha = -4.95^\circ$; $C_L = -0.556$				$\alpha = 1.05^\circ$; $C_L = 0.075$				$\alpha = 4.88^\circ$; $C_L = 0.511$			
153.0	60.22	0.075	0.070	-0.063	-0.020	0.060	0.040	-0.045	0.002	0.042	0.024	-0.025	0.023
154.4	60.80		.106	-.010	.023		.088	.003	.034		.074	.017	.053
156.1	61.46	.164	.132	.035	.054	.142	.124	.046	.057	.133	.114	.056	.074
157.6	62.04	.196	.152	.063	.087	.172	.148	.075	.082	.166	.141	.084	.100
159.2	62.67	.162	.124	.086	.064	.156	.126	.094	.069	.147	.119	.097	.079
160.8	63.30	.085			.027	.095			.047	.090			.046
		$\alpha = -4.00^\circ$; $C_L = -0.475$				$\alpha = 1.87^\circ$; $C_L = 0.171$				$\alpha = 5.96^\circ$; $C_L = 0.637$			
153.0	60.22	0.074	0.060	-0.058	-0.009	0.058	0.040	-0.038	0.003	0.035	0.016	-0.017	0.031
154.4	60.80		.103	-.010	.031		.086	.007	.037		.069	.018	.054
156.1	61.46	.158	.133	.037	.056	.141	.123	.048	.063	.132	.111	.058	.075
157.6	62.04	.190	.151	.068	.090	.170	.145	.079	.089	.161	.140	.086	.103
159.2	62.67	.158	.125	.087	.073	.151	.124	.097	.072	.141	.123	.097	.076
160.8	63.30	.086			.033	.096			.051	.086			.037
		$\alpha = -3.04^\circ$; $C_L = -0.383$				$\alpha = 2.36^\circ$; $C_L = 0.222$				$\alpha = 6.92^\circ$; $C_L = 0.731$			
153.0	60.22	0.072	0.060	-0.056	-0.007	0.046	0.033	-0.036	0.008	0.022	0.008	-0.026	0.029
154.4	60.80		.099	-.007	.031		.080	.006	.039		.060	.016	.056
156.1	61.46	.161	.129	.038	.059	.135	.115	.051	.063	.122	.105	.052	.073
157.6	62.04	.186	.153	.072	.093	.166	.142	.081	.089	.154	.135	.077	.100
159.2	62.67	.160	.125	.089	.071	.150	.118	.092	.069	.135	.112	.088	.075
160.8	63.30	.086			.042	.090			.045	.075			.031
		$\alpha = -2.11^\circ$; $C_L = -0.280$				$\alpha = 2.83^\circ$; $C_L = 0.271$				$\alpha = 8.11^\circ$; $C_L = 0.800$			
153.0	60.22	0.076	0.058	-0.054	-0.008	0.050	0.029	-0.039	0.008	0.020	0.005	-0.014	0.037
154.4	60.80		.099	-.002	.028		.082	.007	.037		.060	.022	.064
156.1	61.46	.158	.131	.040	.059	.138	.118	.052	.068	.122	.102	.056	.078
157.6	62.04	.186	.150	.072	.089	.167	.144	.077	.086	.153	.131	.079	.108
159.2	62.67	.160	.125	.086	.072	.148	.119	.095	.075	.131	.114	.091	.073
160.8	63.30	.091			.048	.089			.052	.078			.027
		$\alpha = -1.16^\circ$; $C_L = -0.174$				$\alpha = 3.39^\circ$; $C_L = 0.331$							
153.0	60.22	0.070	0.056	-0.051	-0.007	0.047	0.033	-0.030	0.008				
154.4	60.80		.098	-.003	.030		.081	.010	.039				
156.1	61.46	.155	.131	.041	.062	.141	.119	.054	.063				
157.6	62.04	.186	.151	.074	.085	.168	.147	.085	.091				
159.2	62.67	.159	.134	.090	.074	.149	.127	.093	.070				
160.8	63.30	.098			.053	.094			.047				
		$\alpha = 0.01^\circ$; $C_L = -0.041$				$\alpha = 4.00^\circ$; $C_L = 0.406$							
153.0	60.22	0.060	0.045	-0.050	-0.005	0.046	0.028	-0.035	0.018				
154.4	60.80		.087	-.009	.026		.074	.010	.046				
156.1	61.46	.143	.130	.040	.057	.134	.116	.053	.069				
157.6	62.04	.175	.147	.071	.080	.166	.139	.083	.094				
159.2	62.67	.152	.124	.088	.070	.146	.122	.095	.074				
160.8	63.30	.091			.052	.087			.043				

TABLE VII - Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5.0^\circ$

(d) M = 0.95

Fuselage station		C_p for -											
cm	in.	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$
		$\alpha = -4.92^\circ$; $C_L = -0.579$				$\alpha = 1.01^\circ$; $C_L = 0.073$				$\alpha = 4.86^\circ$; $C_L = 0.541$			
153.0	60.22	0.074	0.063	-0.128	-0.081	0.055	0.030	-0.068	-0.018	0.035	0.019	-0.035	0.014
154.4	60.80		.099	-.033	.006		.084	-.001	.033		.077	.020	.056
156.1	61.46	.174	.139	.029	.061	.148	.130	.059	.069	.135	.125	.065	.078
157.6	62.04	.210	.162	.070	.101	.184	.154	.089	.099	.174	.154	.093	.109
159.2	62.67	.178	.137	.091	.086	.167	.137	.107	.092	.159	.138	.110	.091
160.8	63.30	.091			.046	.110			.067	.104			.055
		$\alpha = -3.97^\circ$; $C_L = -0.508$				$\alpha = 1.86^\circ$; $C_L = 0.178$				$\alpha = 5.98^\circ$; $C_L = 0.651$			
153.0	60.22	0.068	0.048	-0.120	-0.065	0.051	0.029	-0.059	-0.009	0.041	0.019	-0.024	0.022
154.4	60.80		.090	-.028	.006		.083	.008	.035		.079	.021	.065
156.1	61.46	.166	.135	.031	.055	.149	.127	.055	.067	.143	.126	.067	.088
157.6	62.04	.203	.155	.071	.098	.182	.158	.090	.098	.174	.154	.095	.117
159.2	62.67	.172	.131	.093	.087	.163	.141	.108	.087	.160	.141	.111	.090
160.8	63.30	.092			.049	.114			.070	.104			.053
		$\alpha = -3.03^\circ$; $C_L = -0.415$				$\alpha = 2.29^\circ$; $C_L = 0.227$				$\alpha = 6.58^\circ$; $C_L = 0.707$			
153.0	60.22	0.061	0.045	-0.106	-0.060	0.050	0.029	-0.055	-0.008	0.034	0.015	-0.030	0.022
154.4	60.80		.094	-.022	.014		.085	.010	.039		.076	.012	.057
156.1	61.46	.160	.132	.037	.056	.149	.125	.059	.069	.137	.117	.061	.080
157.6	62.04	.199	.156	.073	.092	.180	.155	.088	.098	.172	.148	.089	.114
159.2	62.67	.167	.134	.098	.087	.170	.142	.107	.088	.153	.131	.102	.084
160.8	63.30	.099			.056	.111			.065	.098			.048
		$\alpha = -2.07^\circ$; $C_L = -0.310$				$\alpha = 2.87^\circ$; $C_L = 0.295$				$\alpha = 8.10^\circ$; $C_L = 0.831$			
153.0	60.22	0.062	0.038	-0.105	-0.053	0.054	0.040	-0.041	0.005	0.030	0.012	-0.024	0.033
154.4	60.80		.092	-.020	.014		.091	.016	.042		.069	.017	.068
156.1	61.46	.159	.130	.038	.056	.149	.134	.067	.073	.131	.117	.057	.079
157.6	62.04	.191	.155	.077	.096	.183	.167	.099	.108	.165	.144	.090	.113
159.2	62.67	.168	.136	.100	.089	.173	.146	.118	.092	.149	.129	.102	.086
160.8	63.30	.102			.063	.114			.072	.095			.039
		$\alpha = -1.12^\circ$; $C_L = -0.192$				$\alpha = 3.38^\circ$; $C_L = 0.357$				$\alpha = 8.92^\circ$; $C_L = 0.890$			
153.0	60.22	0.059	0.043	-0.082	-0.041	0.050	0.026	-0.043	0.003	0.015	0.006	-0.022	0.039
154.4	60.80		.093	-.010	.023		.088	.014	.044		.065	.014	.065
156.1	61.46	.157	.131	.048	.062	.143	.129	.064	.079	.131	.117	.055	.089
157.6	62.04	.186	.159	.086	.095	.180	.159	.095	.105	.167	.143	.079	.118
159.2	62.67	.170	.141	.105	.090	.164	.140	.109	.092	.152	.132	.097	.089
160.8	63.30	.108			.072	.110			.068	.096			.033
		$\alpha = 0.01^\circ$; $C_L = -0.049$				$\alpha = 3.92^\circ$; $C_L = 0.425$							
153.0	60.22	0.052	0.038	-0.081	-0.040	0.039	0.025	-0.035	0.002				
154.4	60.80		.088	-.010	.021		.082	.020	.050				
156.1	61.46	.149	.130	.050	.056	.141	.125	.067	.081				
157.6	62.04	.183	.159	.085	.093	.175	.159	.096	.105				
159.2	62.67	.165	.137	.104	.087	.161	.139	.111	.088				
160.8	63.30	.107			.065	.103			.061				

TABLE VII - Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5.0^o$

(e) M = 0.97

Fuselage station		C_p for -											
cm	in.	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$
		$\alpha = -5.03^o$; $C_L = -0.593$				$\alpha = 0.96^o$; $C_L = 0.062$				$\alpha = 5.89^o$; $C_L = 0.643$			
153.0	60.22	0.062	0.047	-0.215	-0.182	0.042	0.020	-0.129	-0.084	0.022	0.009	-0.059	0.004
154.4	60.80		.091	-.095	-.063		.079	-.037	-.003		.079	.013	.054
156.1	61.46	.171	.129	-.006	.022	.151	.128	.039	.058	.141	.132	.068	.089
157.6	62.04	.214	.150	.043	.082	.186	.160	.091	.097	.175	.158	.103	.122
159.2	62.67	.173	.127	.082	.083	.174	.146	.114	.097	.168	.151	.120	.102
160.8	63.30	.090			.047	.127			.084	.117			.058
		$\alpha = -3.98^o$; $C_L = -0.508$				$\alpha = 1.89^o$; $C_L = 0.184$				$\alpha = 6.86^o$; $C_L = 0.734$			
153.0	60.22	0.056	0.036	-0.198	-0.165	0.030	0.012	-0.142	-0.083	0.022	0.011	-0.050	0.012
154.4	60.80		.087	-.081	-.059		.079	-.031	-.005		.074	.012	.053
156.1	61.46	.163	.126	.005	.024	.150	.122	.047	.054	.138	.127	.064	.085
157.6	62.04	.205	.151	.053	.080	.183	.161	.089	.099	.172	.158	.097	.121
159.2	62.67	.172	.132	.088	.084	.173	.145	.121	.093	.168	.146	.117	.094
160.8	63.30	.094			.053	.122			.082	.109			.057
		$\alpha = -3.00^o$; $C_L = -0.415$				$\alpha = 2.35^o$; $C_L = 0.240$				$\alpha = 8.08^o$; $C_L = 0.832$			
153.0	60.22	0.055	0.037	-0.181	-0.147	0.029	0.016	-0.117	-0.059	0.022	0.011	-0.029	0.029
154.4	60.80		.088	-.067	-.052		.075	-.022	.014		.079	.017	.063
156.1	61.46	.161	.132	.013	.029	.146	.124	.052	.063	.139	.121	.063	.092
157.6	62.04	.202	.154	.061	.085	.179	.157	.095	.103	.178	.158	.095	.128
159.2	62.67	.174	.137	.097	.085	.173	.146	.119	.095	.162	.149	.105	.100
160.8	63.30	.102			.062	.122			.076	.111			.058
		$\alpha = -2.06^o$; $C_L = -0.315$				$\alpha = 3.37^o$; $C_L = 0.367$				$\alpha = 8.87^o$; $C_L = 0.891$			
153.0	60.22	0.052	0.029	-0.178	-0.128	0.033	0.020	-0.116	-0.062	0.010	0.002	-0.028	0.035
154.4	60.80		.079	-.068	-.039		.083	-.014	.019		.069	.017	.066
156.1	61.46	.155	.127	.017	.029	.143	.130	.058	.069	.133	.118	.060	.089
157.6	62.04	.192	.154	.070	.088	.180	.163	.101	.107	.172	.150	.091	.128
159.2	62.67	.170	.136	.101	.088	.173	.150	.122	.103	.162	.146	.102	.096
160.8	63.30	.108			.068	.119			.076	.110			.044
		$\alpha = -1.09^o$; $C_L = -0.198$				$\alpha = 3.93^o$; $C_L = 0.440$				$\alpha = 9.62^o$; $C_L = 0.947$			
153.0	60.22	0.047	0.024	-0.172	-0.122	0.022	0.007	-0.111	-0.047	0.005	0.002	-0.011	0.039
154.4	60.80		.081	-.059	-.034		.075	-.016	.026		.069	.018	.073
156.1	61.46	.151	.128	.029	.039	.141	.127	.064	.072	.134	.121	.058	.096
157.6	62.04	.194	.160	.074	.087	.178	.161	.100	.111	.171	.148	.088	.128
159.2	62.67	.175	.142	.104	.087	.172	.147	.119	.099	.159	.143	.102	.100
160.8	63.30	.115			.076	.117			.073	.111			.048
		$\alpha = 0.04^o$; $C_L = -0.052$				$\alpha = 4.84^o$; $C_L = 0.537$				$\alpha = 10.08^o$; $C_L = 0.973$			
153.0	60.22	0.038	0.019	-0.157	-0.102	0.022	0.004	-0.080	-0.018	0.005	-0.002	-0.018	0.044
154.4	60.80		.083	-.055	-.025		.076	.008	.042		.071	.023	.076
156.1	61.46	.153	.126	.039	.043	.139	.129	.067	.082	.130	.117	.060	.091
157.6	62.04	.190	.162	.082	.090	.178	.161	.104	.118	.166	.149	.086	.131
159.2	62.67	.174	.145	.111	.092	.169	.150	.122	.102	.155	.138	.103	.095
160.8	63.30	.122			.078	.118			.067	.106			.043

[redacted]

TABLE VII - Concluded

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITHOUT FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5.0^{\circ}$

(f) M = 0.99

Fuselage station		C_p for -											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha \approx -4.96^{\circ}$; $C_L = -0.599$				$\alpha = 1.83^{\circ}$; $C_L = 0.171$				$\alpha = 6.82^{\circ}$; $C_L = 0.727$			
153.0	60.22	0.056	0.029	-0.269	-0.225	-0.043	-0.106	-0.201	-0.132	0.007	0.005	-0.110	-0.050
154.4	60.80		.083	-.145	-.140		.024	-.127	-.086		.081	-.018	.019
156.1	61.46	.180	.132	-.036	-.036	.118	.098	-.014	-.015	.143	.134	.061	.084
157.6	62.04	.232	.153	.028	.054	.165	.138	.057	.055	.182	.167	.102	.130
159.2	62.67	.187	.130	.073	.068	.160	.130	.101	.067	.177	.159	.126	.115
160.8	63.30	.095			.054	.113			.072	.131			.073
		$\alpha \approx -3.94^{\circ}$; $C_L = -0.526$				$\alpha = 2.37^{\circ}$; $C_L = 0.247$				$\alpha = 7.94^{\circ}$; $C_L = 0.824$			
153.0	60.22	0.048	0.022	-0.267	-0.207	-0.030	-0.085	-0.200	-0.129	0.011	0.001	-0.115	-0.039
154.4	60.80		.077	-.151	-.133		.035	-.109	-.075		.074	-.021	.017
156.1	61.46	.175	.125	-.034	-.042	.123	.105	-.003	-.004	.144	.125	.057	.087
157.6	62.04	.223	.154	.029	.042	.169	.143	.067	.064	.180	.161	.101	.137
159.2	62.67	.183	.132	.072	.065	.164	.142	.103	.083	.176	.156	.119	.109
160.8	63.30	.095			.049	.115			.079	.129			.069
		$\alpha \approx -3.00^{\circ}$; $C_L = -0.431$				$\alpha = 2.88^{\circ}$; $C_L = 0.311$				$\alpha = 9.06^{\circ}$; $C_L = 0.901$			
153.0	60.22	0.038	0.015	-0.242	-0.191	-0.044	-0.108	-0.206	-0.126	-0.007	-0.002	-0.084	-0.029
154.4	60.80		.072	-.133	-.118		.022	-.129	-.081		.072	-.009	.042
156.1	61.46	.157	.116	-.022	-.029	.122	.100	-.015	-.017	.135	.124	.058	.093
157.6	62.04	.204	.150	.044	.043	.163	.142	.060	.056	.183	.158	.098	.136
159.2	62.67	.172	.125	.084	.069	.158	.139	.100	.075	.176	.160	.119	.114
160.8	63.30	.091			.054	.117			.075	.130			.063
		$\alpha = -2.05^{\circ}$; $C_L = -0.336$				$\alpha = 3.39^{\circ}$; $C_L = 0.373$				$\alpha = 10.05^{\circ}$; $C_L = 0.978$			
153.0	60.22	0.016	-0.003	-0.240	-0.180	-0.051	-0.115	-0.198	-0.118	-0.010	-0.017	-0.088	-0.015
154.4	60.80		.064	-.135	-.117		.017	-.116	-.072		.068	-.007	.037
156.1	61.46	.151	.115	-.029	-.046	.114	.100	-.003	-.006	.138	.117	.059	.091
157.6	62.04	.198	.150	.042	.039	.164	.144	.066	.063	.180	.154	.094	.136
159.2	62.67	.172	.133	.084	.061	.159	.143	.107	.082	.176	.153	.114	.107
160.8	63.30	.103			.062	.116			.074	.130			.062
		$\alpha \approx -1.08^{\circ}$; $C_L = -0.220$				$\alpha = 3.92^{\circ}$; $C_L = 0.435$				$\alpha = 11.94^{\circ}$; $C_L = 1.089$			
153.0	60.22	0.005	-0.038	-0.238	-0.168	-0.044	-0.098	-0.182	-0.102	-0.028	-0.023	-0.043	0.021
154.4	60.80		.058	-.137	-.106		.032	-.099	-.051		.051	.020	.072
156.1	61.46	.139	.111	-.023	-.034	.118	.105	.014	.016	.134	.106	.059	.099
157.6	62.04	.190	.149	.040	.038	.163	.147	.076	.081	.188	.140	.090	.146
159.2	62.67	.175	.140	.088	.063	.161	.140	.112	.092	.189	.153	.105	.114
160.8	63.30	.112			.070	.117			.078	.139			.060
		$\alpha \approx -0.02^{\circ}$; $C_L = -0.079$				$\alpha = 4.91^{\circ}$; $C_L = 0.539$							
153.0	60.22	-0.014	-0.068	-0.223	-0.154	-0.023	-0.062	-0.134	-0.070				
154.4	60.80		.043	-.137	-.101		.051	-.049	-.017				
156.1	61.46	.136	.110	-.025	-.032	.128	.118	.044	.052				
157.6	62.04	.179	.147	.051	.039	.172	.160	.098	.105				
159.2	62.67	.171	.134	.089	.066	.171	.152	.120	.100				
160.8	63.30	.115			.072	.126			.076				
		$\alpha = 0.93^{\circ}$; $C_L \approx 0.048$				$\alpha = 5.91^{\circ}$; $C_L = 0.635$							
153.0	60.22	-0.024	-0.065	-0.203	-0.135	-0.001	-0.011	-0.106	-0.040				
154.4	60.80		.046	-.115	-.083		.067	-.009	.030				
156.1	61.46	.129	.107	.000	-.010	.139	.130	.069	.083				
157.6	62.04	.176	.150	.057	.053	.179	.166	.106	.127				
159.2	62.67	.170	.141	.104	.077	.173	.158	.126	.111				
160.8	63.30	.123			.081	.129			.072				

[redacted]

TABLE VIII

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(a) M = 0.25

Fuselage station		C_p for –											
cm	in.	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$
		$\alpha = -5.08^\circ$; $C_L = -0.411$				$\alpha = 1.97^\circ$; $C_L = 0.216$				$\alpha = 6.98^\circ$; $C_L = 0.638$			
153.0	60.22	0.065	0.065	-0.016	0.024	0.045	0.035	0.016	0.043	0.028	0.007	0.042	0.069
154.4	60.80		.085	.010	.042		.056	.032	.056		.040	.055	.079
156.1	61.46	.119	.096	.041	.056	.099	.079	.051	.065	.083	.062	.072	.080
157.6	62.04	.142	.101	.061	.074	.123	.089	.065	.076	.105	.081	.082	.099
159.2	62.67		.081	.064	.049		.075	.072	.052		.063	.081	.061
160.8	63.30	.052			.011	.051			.028	.042			.020
		$\alpha = -4.12^\circ$; $C_L = -0.329$				$\alpha = 2.48^\circ$; $C_L = 0.260$				$\alpha = 8.00^\circ$; $C_L = 0.714$			
153.0	60.22	0.067	0.060	-0.006	0.025	0.043	0.031	0.013	0.049	0.023	0.006	0.043	0.079
154.4	60.80		.077	.014	.042		.062	.028	.061		.041	.057	.089
156.1	61.46	.121	.096	.034	.058	.100	.082	.055	.066	.081	.068	.076	.090
157.6	62.04	.140	.102	.053	.074	.120	.091	.070	.076	.106	.078	.088	.107
159.2	62.67		.082	.066	.049		.071	.075	.049		.062	.090	.066
160.8	63.30	.052			.011	.054			.021	.049			.024
		$\alpha = -3.10^\circ$; $C_L = -0.234$				$\alpha = 2.91^\circ$; $C_L = 0.299$				$\alpha = 9.02^\circ$; $C_L = 0.791$			
153.0	60.22	0.061	0.057	-0.008	0.031	0.044	0.029	0.022	0.048	0.020	0.000	0.045	0.072
154.4	60.80		.079	.018	.051		.054	.037	.059		.028	.058	.082
156.1	61.46	.108	.096	.035	.059	.102	.076	.055	.063	.079	.054	.070	.087
157.6	62.04	.131	.100	.056	.074	.122	.087	.065	.077	.101	.077	.081	.102
159.2	62.67		.078	.064	.048		.071	.074	.055		.059	.077	.067
160.8	63.30	.050			.014	.051			.029	.038			.023
		$\alpha = -2.08^\circ$; $C_L = -0.142$				$\alpha = 3.43^\circ$; $C_L = 0.343$				$\alpha = 10.02^\circ$; $C_L = 0.868$			
153.0	60.22	0.054	0.050	-0.004	0.035	0.044	0.030	0.024	0.048	0.017	-0.004	0.053	0.080
154.4	60.80		.071	.017	.045		.056	.041	.069		.023	.062	.093
156.1	61.46	.108	.089	.040	.057	.099	.077	.057	.071	.077	.051	.069	.098
157.6	62.04	.131	.098	.058	.068	.119	.090	.070	.083	.102	.070	.083	.108
159.2	62.67		.074	.069	.052		.072	.076	.056		.055	.079	.070
160.8	63.30	.057			.017	.050			.025	.043			.023
		$\alpha = -1.09^\circ$; $C_L = -0.050$				$\alpha = 3.99^\circ$; $C_L = 0.391$				$\alpha = 11.03^\circ$; $C_L = 0.946$			
153.0	60.22	0.052	0.047	-0.003	0.038	0.039	0.022	0.024	0.046	0.005	-0.018	0.049	0.084
154.4	60.80		.069	.020	.052		.046	.038	.062		.015	.058	.093
156.1	61.46	.109	.082	.045	.058	.092	.073	.057	.071	.069	.040	.070	.093
157.6	62.04	.131	.092	.061	.072	.115	.087	.070	.083	.090	.057	.082	.107
159.2	62.67		.071	.067	.049		.068	.072	.056		.044	.077	.066
160.8	63.30	.055			.024	.050			.024	.030			.014
		$\alpha = -0.05^\circ$; $C_L = 0.043$				$\alpha = 4.97^\circ$; $C_L = 0.472$							
153.0	60.22	0.055	0.044	0.008	0.041	0.033	0.024	0.030	0.057				
154.4	60.80		.064	.028	.050		.050	.042	.067				
156.1	61.46	.105	.086	.049	.065	.089	.073	.061	.072				
157.6	62.04	.124	.100	.063	.075	.110	.088	.069	.090				
159.2	62.67		.075	.067	.052		.065	.074	.062				
160.8	63.30	.051			.026	.051			.026				
		$\alpha = 0.98^\circ$; $C_L = 0.132$				$\alpha = 6.03^\circ$; $C_L = 0.559$							
153.0	60.22	0.051	0.036	0.013	0.038	0.030	0.007	0.035	0.060				
154.4	60.80		.063	.031	.049		.036	.049	.067				
156.1	61.46	.100	.084	.053	.059	.081	.059	.066	.072				
157.6	62.04	.119	.093	.066	.073	.104	.079	.074	.085				
159.2	62.67		.073	.069	.052		.065	.076	.058				
160.8	63.30	.052			.027	.040			.023				

TABLE VIII – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(b) M = 0.50

Fuselage station		C_p for –											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -5.10^{\circ}$; $C_L = -0.431$				$\alpha = 2.02^{\circ}$; $C_L = 0.220$				$\alpha = 6.99^{\circ}$; $C_L = 0.649$			
153.0	60.22	0.074	0.069	-0.015	0.023	0.044	0.040	0.013	0.045	0.027	0.010	0.036	0.066
154.4	60.80		.087	.013	.047		.064	.032	.064		.043	.054	.082
156.1	61.46	.129	.108	.035	.060	.106	.087	.058	.070	.095	.067	.068	.085
157.6	62.04	.152	.113	.061	.083	.131	.099	.075	.082	.113	.088	.085	.108
159.2	62.67		.088	.070	.052		.081	.078	.061		.067	.088	.067
160.8	63.30	.055			.010	.062			.029	.053			.020
		$\alpha = -4.08^{\circ}$; $C_L = -0.339$				$\alpha = 2.49^{\circ}$; $C_L = 0.259$				$\alpha = 8.00^{\circ}$; $C_L = 0.730$			
153.0	60.22	0.073	0.062	-0.014	0.028	0.046	0.038	0.013	0.046	0.020	0.003	0.047	0.073
154.4	60.80		.084	.010	.046		.055	.035	.061		.032	.057	.091
156.1	61.46	.127	.100	.041	.059	.103	.083	.059	.069	.085	.065	.075	.091
157.6	62.04	.145	.105	.062	.081	.131	.096	.072	.083	.110	.079	.084	.103
159.2	62.67		.084	.070	.055		.076	.077	.057		.060	.087	.067
160.8	63.30	.058			.015	.058			.027	.046			.018
		$\alpha = -3.08^{\circ}$; $C_L = -0.247$				$\alpha = 2.98^{\circ}$; $C_L = 0.305$				$\alpha = 9.02^{\circ}$; $C_L = 0.809$			
153.0	60.22	0.068	0.057	-0.003	0.025	0.042	0.031	0.013	0.049	0.019	-0.005	0.048	0.081
154.4	60.80		.081	.018	.048		.057	.035	.062		.026	.060	.094
156.1	61.46	.122	.100	.045	.062	.102	.082	.060	.073	.084	.060	.078	.100
157.6	62.04	.146	.109	.061	.078	.126	.096	.073	.083	.108	.080	.091	.109
159.2	62.67		.087	.074	.053		.076	.079	.055		.066	.091	.071
160.8	63.30	.061			.019	.063			.023	.047			.019
		$\alpha = -2.08^{\circ}$; $C_L = -0.154$				$\alpha = 3.48^{\circ}$; $C_L = 0.351$				$\alpha = 10.00^{\circ}$; $C_L = 0.884$			
153.0	60.22	0.055	0.049	-0.009	0.030	0.037	0.029	0.021	0.051	0.014	-0.008	0.049	0.083
154.4	60.80		.075	.018	.049		.053	.034	.063		.021	.066	.097
156.1	61.46	.115	.095	.042	.062	.100	.079	.059	.069	.082	.056	.077	.100
157.6	62.04	.138	.102	.061	.074	.124	.096	.072	.084	.109	.073	.091	.115
159.2	62.67		.084	.068	.054		.072	.076	.060		.063	.088	.069
160.8	63.30	.055			.026	.056			.027	.043			.019
		$\alpha = -1.08^{\circ}$; $C_L = -0.578$				$\alpha = 4.00^{\circ}$; $C_L = 0.396$				$\alpha = 11.07^{\circ}$; $C_L = 0.952$			
153.0	60.22	0.051	0.047	-0.003	0.038	0.041	0.025	0.020	0.053	0.004	-0.012	0.052	0.091
154.4	60.80		.069	.024	.053		.054	.042	.067		.012	.059	.098
156.1	61.46	.114	.093	.042	.061	.101	.080	.062	.076	.078	.054	.084	.104
157.6	62.04	.137	.104	.064	.076	.124	.093	.080	.092	.110	.075	.088	.117
159.2	62.67		.083	.070	.052		.074	.079	.061		.063	.090	.070
160.8	63.30	.061			.027	.061			.031	.048			.017
		$\alpha = 0^{\circ}$; $C_L = 0.394$				$\alpha = 4.97^{\circ}$; $C_L = 0.481$							
153.0	60.22	0.051	0.042	0.000	0.034	0.036	0.020	0.025	0.063				
154.4	60.80		.066	.025	.053		.051	.045	.074				
156.1	61.46	.108	.087	.049	.060	.101	.076	.063	.079				
157.6	62.04	.134	.102	.066	.075	.120	.091	.081	.090				
159.2	62.67		.080	.073	.054		.077	.084	.059				
160.8	63.30	.057			.026	.058			.021				
		$\alpha = 1.03^{\circ}$; $C_L = 0.130$				$\alpha = 6.04^{\circ}$; $C_L = 0.572$							
153.0	60.22	0.048	0.038	0.010	0.041	0.030	0.012	0.026	0.063				
154.4	60.80		.066	.029	.057		.041	.047	.074				
156.1	61.46	.106	.089	.052	.064	.089	.070	.069	.082				
157.6	62.04	.128	.103	.068	.077	.114	.083	.085	.092				
159.2	62.67		.078	.073	.055		.067	.083	.060				
160.8	63.30	.055			.029	.053			.021				

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TABLE VIII – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(c) M = 0.80

Fuselage station		C_p for –											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -4.96^{\circ}$; $C_L = -0.455$				$\alpha = 2.01^{\circ}$; $C_L = 0.231$				$\alpha = 7.08^{\circ}$; $C_L = 0.719$			
153.0	60.22	0.070	0.063	-0.023	0.019	0.043	0.036	0.013	0.044	0.020	-0.001	0.033	0.069
154.4	60.80		.094	.011	.045		.069	.035	.066		.049	.056	.085
156.1	61.46	.145	.119	.049	.064	.121	.103	.068	.076	.101	.081	.082	.094
157.6	62.04	.174	.130	.077	.094	.149	.123	.085	.100	.135	.110	.096	.115
159.2	62.67		.100	.090	.068		.104	.097	.070		.101	.104	.075
160.8	63.30	.063			.020	.083			.043	.072			.024
		$\alpha = -3.98^{\circ}$; $C_L = -0.372$				$\alpha = 2.47^{\circ}$; $C_L = 0.276$				$\alpha = 8.14^{\circ}$; $C_L = 0.792$			
153.0	60.22	0.072	0.061	-0.015	0.021	0.048	0.034	0.011	0.046	0.016	-0.012	0.035	0.082
154.4	60.80		.090	.014	.048		.068	.036	.066		.037	.061	.091
156.1	61.46	.150	.117	.055	.079	.124	.104	.069	.076	.103	.078	.083	.097
157.6	62.04	.175	.134	.074	.099	.151	.123	.090	.099	.122	.110	.100	.122
159.2	62.67		.109	.089	.078		.096	.095	.073		.086	.106	.073
160.8	63.30	.071			.025	.076			.043	.068			.030
		$\alpha = -3.00^{\circ}$; $C_L = -0.278$				$\alpha = 2.94^{\circ}$; $C_L = 0.321$				$\alpha = 9.17^{\circ}$; $C_L = 0.857$			
153.0	60.22	0.064	0.055	-0.014	0.026	0.040	0.027	0.009	0.053	0.001	-0.012	0.044	0.076
154.4	60.80		.088	.016	.049		.065	.038	.064		.036	.069	.098
156.1	61.46	.140	.116	.051	.068	.120	.099	.074	.082	.096	.075	.092	.103
157.6	62.04	.169	.131	.076	.091	.151	.117	.092	.099	.125	.101	.105	.125
159.2	62.67		.110	.087	.068		.095	.097	.074		.087	.106	.083
160.8	63.30	.070			.030	.080			.040	.063			.024
		$\alpha = -2.02^{\circ}$; $C_L = -0.178$				$\alpha = 3.45^{\circ}$; $C_L = 0.370$				$\alpha = 10.22^{\circ}$; $C_L = 0.927$			
153.0	60.22	0.067	0.053	-0.009	0.025	0.044	0.031	0.016	0.054	-0.006	-0.020	0.049	0.090
154.4	60.80		.084	.021	.054		.064	.047	.071		.031	.071	.098
156.1	61.46	.140	.109	.056	.072	.119	.097	.079	.083	.084	.070	.091	.104
157.6	62.04	.161	.130	.078	.090	.149	.117	.092	.106	.111	.099	.102	.125
159.2	62.67		.104	.090	.066		.097	.103	.077		.089	.099	.077
160.8	63.30	.072			.034	.077			.044	.056			.023
		$\alpha = -1.03^{\circ}$; $C_L = -0.075$				$\alpha = 4.02^{\circ}$; $C_L = 0.427$				$\alpha = 11.22^{\circ}$; $C_L = 0.990$			
153.0	60.22	0.061	0.049	-0.001	0.028	0.037	0.026	0.015	0.052	-0.014	-0.036	0.051	0.090
154.4	60.80		.079	.029	.050		.062	.043	.070		.018	.071	.112
156.1	61.46	.131	.108	.057	.066	.113	.097	.069	.085	.087	.057	.086	.114
157.6	62.04	.159	.126	.083	.089	.142	.122	.088	.100	.125	.090	.104	.130
159.2	62.67		.107	.092	.074		.100	.097	.072		.082	.102	.082
160.8	63.30	.075			.045	.075			.038	.052			.017
		$\alpha = 0.03^{\circ}$; $C_L = -0.034$				$\alpha = 4.97^{\circ}$; $C_L = 0.524$							
153.0	60.22	0.047	0.043	-0.002	0.033	0.033	0.020	0.022	0.055				
154.4	60.80		.076	.031	.057		.054	.050	.070				
156.1	61.46	.123	.109	.062	.072	.114	.088	.080	.083				
157.6	62.04	.153	.128	.078	.090	.145	.117	.101	.106				
159.2	62.67		.103	.090	.069		.091	.097	.079				
160.8	63.30	.075			.046	.074			.030				
		$\alpha = 1.02^{\circ}$; $C_L = 0.130$				$\alpha = 6.06^{\circ}$; $C_L = 0.628$							
153.0	60.22	0.048	0.040	0.005	0.044	0.032	0.012	0.029	0.063				
154.4	60.80		.075	.037	.062		.055	.053	.085				
156.1	61.46	.127	.106	.063	.077	.108	.089	.078	.097				
157.6	62.04	.152	.125	.086	.091	.140	.113	.100	.116				
159.2	62.67		.106	.094	.067		.089	.102	.076				
160.8	63.30	.079			.044	.073			.033				

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TABLE VIII – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^\circ$

(d) M = 0.90

Fuselage station		C_p for –											
cm	in.	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$	$\theta = 8^\circ$	$\theta = 46^\circ$	$\theta = 136^\circ$	$\theta = 180^\circ$
		$\alpha = -4.90^\circ$; $C_L = -0.482$				$\alpha = 1.05^\circ$; $C_L = 0.135$				$\alpha = 4.94^\circ$; $C_L = 0.580$			
153.0	60.22	0.071	0.059	-0.021	0.019	0.049	0.027	0.005	0.036	0.030	0.010	0.019	0.057
154.4	60.80		.097	.022	.057		.076	.038	.065		.063	.053	.080
156.1	61.46	.156	.132	.062	.082	.133	.117	.075	.083	.118	.102	.083	.096
157.6	62.04	.196	.146	.093	.110	.164	.141	.098	.102	.151	.133	.103	.116
159.2	62.67		.120	.102	.085		.123	.109	.090		.117	.110	.089
160.8	63.30	.080			.038	.094			.061	.089			.054
		$\alpha = -3.82^\circ$; $C_L = -0.389$				$\alpha = 2.01^\circ$; $C_L = 0.239$				$\alpha = 6.06^\circ$; $C_L = 0.706$			
153.0	60.22	0.064	0.054	-0.021	0.018	0.042	0.025	0.002	0.040	0.022	-0.000	0.029	0.069
154.4	60.80		.092	.021	.054		.071	.039	.066		.056	.056	.087
156.1	61.46	.155	.130	.062	.079	.130	.113	.077	.085	.115	.103	.091	.098
157.6	62.04	.190	.144	.089	.109	.160	.140	.097	.106	.150	.134	.108	.126
159.2	62.67		.117	.103	.088		.118	.104	.087		.112	.109	.091
160.8	63.30	.082			.039	.096			.060	.093			.051
		$\alpha = -2.96^\circ$; $C_L = -0.305$				$\alpha = 2.45^\circ$; $C_L = 0.285$				$\alpha = 7.15^\circ$; $C_L = 0.801$			
153.0	60.22	0.066	0.052	-0.015	0.022	0.047	0.027	0.004	0.043	0.017	-0.011	0.031	0.071
154.4	60.80		.094	.024	.056		.073	.040	.066		.048	.060	.090
156.1	61.46	.150	.128	.063	.075	.137	.114	.075	.085	.109	.095	.091	.101
157.6	62.04	.190	.148	.088	.109	.161	.141	.098	.109	.135	.124	.108	.123
159.2	62.67		.122	.107	.088		.119	.107	.087		.106	.115	.087
160.8	63.30	.088			.054	.097			.060	.087			.042
		$\alpha = -2.00^\circ$; $C_L = -0.203$				$\alpha = 2.94^\circ$; $C_L = 0.341$				$\alpha = 8.17^\circ$; $C_L = 0.863$			
153.0	60.22	0.064	0.045	-0.012	0.027	0.035	0.019	0.008	0.046	0.002	-0.017	0.034	0.069
154.4	60.80		.090	.030	.054		.070	.040	.068		.043	.065	.093
156.1	61.46	.150	.124	.068	.079	.124	.110	.072	.084	.102	.090	.092	.106
157.6	62.04	.180	.144	.094	.104	.154	.132	.098	.105	.135	.125	.108	.128
159.2	62.67		.125	.104	.088		.116	.108	.087		.108	.115	.090
160.8	63.30	.093			.059	.096			.058	.085			.041
		$\alpha = -1.04^\circ$; $C_L = -0.094$				$\alpha = 3.47^\circ$; $C_L = 0.403$							
153.0	60.22	0.057	0.041	-0.009	0.027	0.030	0.013	0.010	0.046				
154.4	60.80		.086	.033	.056		.060	.043	.069				
156.1	61.46	.142	.124	.070	.079	.125	.107	.076	.087				
157.6	62.04	.176	.145	.089	.102	.153	.132	.100	.109				
159.2	62.67		.121	.108	.089		.112	.108	.080				
160.8	63.30	.093			.059	.090			.056				
		$\alpha = 0.02^\circ$; $C_L = 0.024$				$\alpha = 3.97^\circ$; $C_L = 0.461$							
153.0	60.22	0.052	0.036	-0.005	0.030	0.033	0.014	0.012	0.051				
154.4	60.80		.078	.033	.056		.066	.050	.073				
156.1	61.46	.138	.119	.066	.081	.126	.107	.080	.087				
157.6	62.04	.166	.136	.094	.099	.155	.133	.104	.110				
159.2	62.67		.118	.105	.087		.115	.110	.087				
160.8	63.30	.097			.062	.095			.049				

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TABLE VIII - Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(e) $M = 0.95$

Fuselage station		C_p for -											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -4.93^{\circ}$; $C_L = -0.524$				$\alpha = 0.99^{\circ}$; $C_L = 0.140$				$\alpha = 4.92^{\circ}$; $C_L = 0.606$			
153.0	60.22	0.072	0.053	-0.039	0.007	0.047	0.030	-0.002	0.033	0.028	0.005	0.021	0.057
154.4	60.80		.100	.022	.055		.085	.041	.062		.068	.056	.088
156.1	61.46	.174	.139	.071	.086	.146	.127	.085	.092	.132	.120	.093	.101
157.6	62.04	.217	.160	.105	.125	.180	.160	.113	.112	.163	.150	.117	.128
159.2	62.67		.142	.122	.106		.136	.126	.098		.132	.124	.103
160.8	63.30	.096			.053	.115			.077	.107			.064
		$\alpha = -3.97^{\circ}$; $C_L = -0.440$				$\alpha = 1.96^{\circ}$; $C_L = 0.253$				$\alpha = 5.95^{\circ}$; $C_L = 0.711$			
153.0	60.22	0.066	0.050	-0.024	0.012	0.045	0.020	0.003	0.040	0.016	-0.004	0.027	0.066
154.4	60.80		.100	.025	.054		.080	.046	.072		.063	.062	.092
156.1	61.46	.169	.141	.075	.087	.144	.126	.085	.095	.124	.114	.092	.106
157.6	62.04	.214	.158	.105	.122	.173	.155	.111	.118	.160	.149	.117	.131
159.2	62.67		.135	.120	.101		.136	.126	.103		.131	.128	.104
160.8	63.30	.097			.060	.114			.080	.111			.061
		$\alpha = -3.03^{\circ}$; $C_L = -0.346$				$\alpha = 2.41^{\circ}$; $C_L = 0.306$				$\alpha = 7.03^{\circ}$; $C_L = 0.813$			
153.0	60.22	0.063	0.045	-0.025	0.014	0.039	0.022	0.005	0.036	0.007	-0.010	0.035	0.075
154.4	60.80		.094	.030	.056		.078	.046	.068		.060	.065	.100
156.1	61.46	.164	.135	.074	.090	.140	.125	.090	.088	.123	.113	.102	.116
157.6	62.04	.203	.159	.103	.117	.169	.153	.112	.112	.156	.152	.121	.138
159.2	62.67		.136	.122	.106		.136	.122	.097		.132	.130	.109
160.8	63.30	.101			.065	.112			.083	.108			.059
		$\alpha = -2.05^{\circ}$; $C_L = -0.230$				$\alpha = 2.92^{\circ}$; $C_L = 0.369$				$\alpha = 8.15^{\circ}$; $C_L = 0.902$			
153.0	60.22	0.059	0.038	-0.021	0.017	0.040	0.020	0.009	0.044	-0.010	-0.029	0.040	0.079
154.4	60.80		.090	.031	.055		.077	.050	.068		.053	.070	.105
156.1	61.46	.159	.131	.073	.087	.140	.126	.088	.091	.120	.109	.094	.117
157.6	62.04	.194	.159	.102	.112	.169	.153	.110	.116	.154	.138	.122	.138
159.2	62.67		.136	.120	.099		.135	.124	.100		.130	.125	.106
160.8	63.30	.105			.075	.114			.075	.103			.051
		$\alpha = -1.10^{\circ}$; $C_L = -0.111$				$\alpha = 3.42^{\circ}$; $C_L = 0.429$				$\alpha = 9.19^{\circ}$; $C_L = 0.969$			
153.0	60.22	0.055	0.036	-0.008	0.022	0.037	0.016	0.009	0.047	-0.023	-0.048	0.042	0.085
154.4	60.80		.090	.036	.058		.072	.048	.076		.043	.071	.112
156.1	61.46	.152	.135	.077	.088	.132	.119	.089	.098	.113	.099	.095	.121
157.6	62.04	.189	.157	.110	.112	.163	.150	.113	.117	.150	.137	.122	.144
159.2	62.67		.137	.124	.101		.131	.124	.098		.127	.128	.107
160.8	63.30	.110			.075	.113			.074	.104			.051
		$\alpha = -0.09^{\circ}$; $C_L = 0.012$				$\alpha = 3.96^{\circ}$; $C_L = 0.495$				$\alpha = 9.58^{\circ}$; $C_L = 0.982$			
153.0	60.22	0.050	0.035	-0.010	0.033	0.030	0.012	0.015	0.053	-0.034	-0.052	0.047	0.085
154.4	60.80		.088	.040	.065		.071	.053	.079		.043	.070	.109
156.1	61.46	.151	.132	.082	.090	.133	.126	.093	.102	.107	.098	.102	.118
157.6	62.04	.186	.156	.108	.116	.167	.154	.118	.128	.147	.135	.118	.141
159.2	62.67		.137	.123	.100		.135	.129	.102		.115	.127	.108
160.8	63.30	.112			.079	.113			.070	.101			.046

[redacted]

TABLE VIII – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(f) M = 0.97

Fuselage station		C_p for –											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -4.99^{\circ}$; $C_L = -0.542$				$\alpha = 1.96^{\circ}$; $C_L = 0.249$				$\alpha = 6.95^{\circ}$; $C_L = 0.809$			
153.0	60.22	0.086	0.056	-0.073	-0.029	0.036	0.018	-0.004	0.031	-0.022	-0.054	0.036	0.077
154.4	60.80		.109	.009	.038		.085	.047	.070		.055	.069	.101
156.1	61.46	.195	.147	.069	.091	.147	.136	.093	.099	.120	.116	.100	.120
157.6	62.04	.242	.172	.110	.133	.182	.165	.119	.126	.158	.156	.126	.144
159.2	62.67		.150	.131	.113		.145	.137	.113		.142	.138	.115
160.8	63.30	.106			.069	.124			.091	.118			.073
		$\alpha = -3.98^{\circ}$; $C_L = -0.455$				$\alpha = 2.43^{\circ}$; $C_L = 0.304$				$\alpha = 8.06^{\circ}$; $C_L = 0.899$			
153.0	60.22	0.070	0.048	-0.075	-0.030	0.033	0.009	0.004	0.039	-0.034	-0.071	0.039	0.082
154.4	60.80		.101	.004	.035		.081	.049	.071		.045	.070	.105
156.1	61.46	.186	.146	.067	.082	.146	.134	.093	.097	.114	.114	.101	.118
157.6	62.04	.232	.166	.103	.122	.180	.163	.121	.124	.152	.147	.125	.147
159.2	62.67		.143	.125	.110		.147	.136	.108		.136	.132	.111
160.8	63.30	.103			.073	.128			.089	.117			.069
		$\alpha = -2.98^{\circ}$; $C_L = -0.352$				$\alpha = 2.93^{\circ}$; $C_L = 0.366$				$\alpha = 9.16^{\circ}$; $C_L = 0.981$			
153.0	60.22	0.061	0.040	-0.065	-0.017	0.020	0.004	0.002	0.045	-0.046	-0.111	0.043	0.088
154.4	60.80		.100	.014	.039		.077	.050	.074		.039	.071	.113
156.1	61.46	.177	.138	.071	.082	.141	.130	.094	.099	.106	.107	.099	.124
157.6	62.04	.222	.165	.104	.119	.177	.161	.121	.127	.153	.142	.123	.142
159.2	62.67		.145	.128	.109		.147	.134	.112		.134	.132	.114
160.8	63.30	.109			.075	.123			.085	.119			.063
		$\alpha = -2.04^{\circ}$; $C_L = -0.241$				$\alpha = 3.44^{\circ}$; $C_L = 0.429$				$\alpha = 10.18^{\circ}$; $C_L = 1.043$			
153.0	60.22	0.054	0.038	-0.042	-0.007	0.018	0.003	0.011	0.047	-0.069	-0.124	0.052	0.092
154.4	60.80		.090	.021	.047		.076	.053	.076		.037	.075	.117
156.1	61.46	.167	.139	.077	.082	.139	.131	.097	.105	.106	.107	.100	.125
157.6	62.04	.211	.166	.111	.118	.171	.164	.122	.127	.149	.144	.125	.150
159.2	62.67		.146	.130	.112		.146	.135	.112		.137	.132	.119
160.8	63.30	.113			.086	.125			.085	.116			.063
		$\alpha = -1.06^{\circ}$; $C_L = -0.114$				$\alpha = 3.95^{\circ}$; $C_L = 0.493$				$\alpha = 11.19^{\circ}$; $C_L = 1.093$			
153.0	60.22	0.051	0.032	-0.029	0.012	0.009	-0.009	0.010	0.050	-0.066	-0.130	0.062	0.106
154.4	60.80		.094	.035	.058		.072	.055	.081		.035	.082	.125
156.1	61.46	.164	.142	.084	.092	.133	.126	.096	.106	.106	.102	.112	.133
157.6	62.04	.203	.169	.118	.122	.170	.159	.123	.130	.148	.139	.130	.158
159.2	62.67		.147	.133	.111		.146	.132	.111		.134	.135	.122
160.8	63.30	.122			.096	.123			.078	.116			.064
		$\alpha = -0.07^{\circ}$; $C_L = 0.010$				$\alpha = 4.98^{\circ}$; $C_L = 0.609$							
153.0	60.22	0.051	0.022	-0.015	0.020	-0.004	-0.018	0.021	0.060				
154.4	60.80		.088	.041	.060		.064	.059	.089				
156.1	61.46	.155	.138	.087	.093	.132	.126	.102	.108				
157.6	62.04	.192	.166	.112	.118	.165	.158	.124	.135				
159.2	62.67		.148	.131	.109		.144	.135	.111				
160.8	63.30	.122			.094	.122			.078				
		$\alpha = 0.97^{\circ}$; $C_L = 0.132$				$\alpha = 5.96^{\circ}$; $C_L = 0.709$							
153.0	60.22	0.042	0.018	-0.011	0.030	-0.012	-0.042	0.029	0.071				
154.4	60.80		.085	.041	.064		.057	.068	.094				
156.1	61.46	.153	.136	.091	.093	.119	.121	.099	.111				
157.6	62.04	.186	.162	.120	.118	.162	.156	.128	.138				
159.2	62.67		.143	.130	.109		.141	.135	.111				
160.8	63.30	.117			.094	.120			.073				

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TABLE VIII - Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5$

(g) M = 0.98

Fuselage station		C_p for –											
cm	in.	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$
		$\alpha = -1.11^o$; $C_L = -0.126$				$\alpha = 1.94^o$; $C_L = 0.252$				$\alpha = 3.41^o$; $C_L = 0.438$			
153.0	60.22	0.050	0.022	-0.047	-0.006	0.012	-0.003	-0.020	0.017	-0.004	-0.037	-0.002	0.041
154.4	60.80		.088	.024	.044		.078	.041	.065		.070	.052	.075
156.1	61.46	.174	.140	.079	.088	.152	.133	.091	.094	.134	.132	.098	.102
157.6	62.04	.215	.168	.115	.119	.186	.164	.120	.123	.174	.166	.128	.131
159.2	62.67		.152	.133	.113		.148	.134	.113		.152	.140	.117
160.8	63.30	.122			.096	.125			.096	.131			.093
		$\alpha = -0.15^o$; $C_L = -0.003$				$\alpha = 2.41^o$; $C_L = 0.312$				$\alpha = 3.93^o$; $C_L = 0.500$			
153.0	60.22	0.040	0.021	-0.028	0.006	0.007	-0.018	-0.019	0.027	-0.005	-0.042	0.005	0.046
154.4	60.80		.088	.027	.050		.075	.040	.066		.065	.055	.082
156.1	61.46	.166	.139	.084	.091	.143	.132	.090	.097	.133	.128	.101	.107
157.6	62.04	.212	.170	.119	.119	.180	.165	.122	.128	.168	.164	.128	.133
159.2	62.67		.154	.138	.114		.151	.138	.116		.151	.140	.117
160.8	63.30	.121			.099	.130			.096	.128			.088
		$\alpha = 0.90^o$; $C_L = 0.126$				$\alpha = 2.92^o$; $C_L = 0.378$				$\alpha = 4.95^o$; $C_L = 0.614$			
153.0	60.22	0.027	0.006	-0.023	0.011	-0.003	-0.029	-0.011	0.032	-0.012	-0.055	0.022	0.060
154.4	60.80		.083	.041	.059		.069	.047	.069		.059	.063	.091
156.1	61.46	.165	.140	.089	.096	.133	.130	.090	.102	.129	.126	.101	.113
157.6	62.04	.208	.172	.123	.126	.174	.164	.121	.125	.166	.163	.127	.138
159.2	62.67		.153	.140	.117		.148	.134	.113		.149	.140	.120
160.8	63.30	.125			.102	.127			.090	.128			.085

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TABLE VIII - Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(h) M = 0.99

Fuselage station		C_p for -											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -4.99^{\circ}$; $C_L = -0.551$				$\alpha = 1.94^{\circ}$; $C_L = 0.247$				$\alpha = 6.96^{\circ}$; $C_L = 0.807$			
153.0	60.22	0.098	0.065	-0.095	-0.056	0.001	-0.038	-0.039	0.002	-0.008	-0.053	0.055	0.092
154.4	60.80		.113	-.009	.016		.067	.024	.055		.065	.088	.120
156.1	61.46	.212	.156	.064	.080	.150	.134	.084	.093	.137	.134	.120	.136
157.6	62.04	.259	.176	.106	.128	.193	.169	.121	.121	.173	.171	.144	.162
159.2	62.67		.158	.138	.124		.156	.139	.114		.161	.154	.135
160.8	63.30	.119			.089	.130			.100	.135			.093
		$\alpha = -3.96^{\circ}$; $C_L = -0.459$				$\alpha = 2.42^{\circ}$; $C_L = 0.308$				$\alpha = 7.99^{\circ}$; $C_L = 0.893$			
153.0	60.22	0.091	0.057	-0.083	-0.052	-0.001	-0.039	-0.033	0.013	-0.016	-0.075	0.056	0.095
154.4	60.80		.106	-.002	.026		.068	.033	.060		.056	.086	.121
156.1	61.46	.203	.151	.066	.078	.150	.134	.092	.097	.130	.127	.117	.135
157.6	62.04	.252	.176	.109	.123	.196	.171	.125	.128	.168	.166	.141	.160
159.2	62.67		.155	.130	.117		.158	.144	.120		.155	.151	.135
160.8	63.30	.116			.087	.133			.105	.134			.087
		$\alpha = -2.98^{\circ}$; $C_L = -0.358$				$\alpha = 2.93^{\circ}$; $C_L = 0.376$				$\alpha = 9.15^{\circ}$; $C_L = 0.982$			
153.0	60.22	0.077	0.049	-0.062	-0.029	0.004	-0.047	-0.008	0.035	-0.035	-0.085	0.059	0.099
154.4	60.80		.102	.011	.040		.070	.046	.072		.048	.078	.121
156.1	61.46	.195	.155	.079	.087	.151	.136	.097	.101	.122	.123	.115	.134
157.6	62.04	.243	.177	.114	.128	.195	.172	.126	.134	.165	.161	.137	.161
159.2	62.67		.158	.142	.122		.158	.142	.120		.153	.148	.131
160.8	63.30	.121			.089	.135			.101	.135			.081
		$\alpha = -1.99^{\circ}$; $C_L = -0.250$				$\alpha = 3.43^{\circ}$; $C_L = 0.440$				$\alpha = 10.18^{\circ}$; $C_L = 1.056$			
153.0	60.22	0.068	0.039	-0.041	-0.006	-0.001	-0.036	0.005	0.046	-0.051	-0.130	0.055	0.100
154.4	60.80		.102	.023	.047		.069	.057	.088		.044	.078	.124
156.1	61.46	.188	.154	.084	.091	.145	.138	.101	.109	.114	.113	.105	.133
157.6	62.04	.233	.179	.119	.125	.191	.171	.131	.136	.159	.152	.129	.158
159.2	62.67		.156	.141	.123		.160	.147	.127		.148	.138	.131
160.8	63.30	.129			.105	.133			.100	.131			.078
		$\alpha = -1.05^{\circ}$; $C_L = -0.134$				$\alpha = 3.94^{\circ}$; $C_L = 0.498$				$\alpha = 11.21^{\circ}$; $C_L = 1.107$			
153.0	60.22	0.055	0.033	-0.027	0.009	0.001	-0.034	0.016	0.058	-0.073	-0.112	0.052	0.099
154.4	60.80		.095	.040	.063		.073	.065	.091		.037	.078	.125
156.1	61.46	.177	.152	.090	.094	.148	.138	.107	.115	.113	.107	.099	.136
157.6	62.04	.227	.178	.121	.131	.190	.175	.134	.141	.160	.147	.125	.158
159.2	62.67		.163	.146	.125		.162	.149	.125		.147	.140	.129
160.8	63.30	.133			.107	.135			.100	.135			.076
		$\alpha = -0.05^{\circ}$; $C_L = -0.004$				$\alpha = 4.95^{\circ}$; $C_L = 0.612$							
153.0	60.22	0.041	0.019	-0.012	0.027	0.005	-0.031	0.041	0.081				
154.4	60.80		.090	.048	.069		.076	.079	.106				
156.1	61.46	.174	.148	.095	.103	.149	.142	.118	.129				
157.1	62.04	.216	.176	.129	.129	.189	.177	.141	.155				
159.2	62.67		.161	.145	.126		.165	.151	.132				
160.8	63.30	.132			.110	.137			.101				
		$\alpha = 0.94^{\circ}$; $C_L = 0.124$				$\alpha = 5.96^{\circ}$; $C_L = 0.713$							
153.0	60.22	0.032	-0.003	-0.016	0.024	-0.000	-0.043	0.047	0.085				
154.4	60.80		.084	.040	.064		.069	.086	.114				
156.1	61.46	.164	.143	.099	.102	.142	.137	.117	.133				
157.1	62.04	.209	.178	.126	.128	.181	.174	.142	.156				
159.2	62.67		.160	.142	.124		.164	.154	.134				
160.8	63.30	.135			.108	.137			.097				

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TABLE VIII - Concluded

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -2.5^{\circ}$

(i) M = 1.00

Fuselage station		C_p for -											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -1.11^{\circ}$; $C_L = -0.142$				$\alpha = 1.96^{\circ}$; $C_L = 0.237$				$\alpha = 3.44^{\circ}$; $C_L = 0.423$			
153.0	60.22	0.017	-0.028	-0.068	-0.030	0.003	-0.043	-0.032	0.007	0.029	-0.006	0.012	0.046
154.4	60.80		.070	.007	.027		.068	.035	.060		.087	.066	.092
156.1	61.46	.165	.136	.072	.080	.156	.139	.099	.103	.159	.148	.119	.127
157.6	62.04	.213	.172	.115	.117	.203	.176	.129	.134	.197	.181	.144	.149
159.2	62.67		.156	.140	.118		.163	.146	.127		.171	.159	.138
160.8	63.30	.133			.110	.141			.110	.148			.118
		$\alpha = -0.11^{\circ}$; $C_L = -0.018$				$\alpha = 2.45^{\circ}$; $C_L = 0.299$				$\alpha = 3.94^{\circ}$; $C_L = 0.486$			
153.0	60.22	-0.013	-0.065	-0.060	-0.025	0.016	-0.030	-0.010	0.033	0.023	-0.005	-0.002	0.042
154.4	60.80		.052	.013	.035		.073	.058	.075		.086	.066	.089
156.1	61.46	.153	.132	.078	.081	.158	.144	.105	.111	.156	.150	.118	.127
157.6	62.04	.200	.169	.115	.118	.204	.177	.138	.141	.194	.185	.146	.155
159.2	62.67		.158	.142	.120		.168	.153	.134		.171	.162	.142
160.8	63.30	.132			.113	.144			.117	.149			.116
		$\alpha = 0.95^{\circ}$; $C_L = 0.114$				$\alpha = 2.96^{\circ}$; $C_L = 0.363$				$\alpha = 4.96^{\circ}$; $C_L = 0.598$			
153.0	60.22	-0.012	-0.077	-0.054	-0.004	0.018	-0.014	0.010	0.041	0.017	-0.013	0.015	0.050
154.4	60.80		.057	.022	.049		.082	.066	.087		.084	.078	.100
156.1	61.46	.151	.132	.085	.091	.157	.144	.112	.116	.149	.144	.120	.133
157.6	62.04	.197	.172	.121	.123	.199	.179	.140	.143	.188	.179	.148	.160
159.2	62.67		.159	.144	.125		.169	.155	.134		.172	.161	.142
160.8	63.30	.138			.114	.143			.115	.151			.114

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TABLE IX

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5^{o}$

(a) M = 0.50

Fuselage station		C_p for –											
cm	in.	$\theta = 8^{o}$	$\theta = 46^{o}$	$\theta = 136^{o}$	$\theta = 180^{o}$	$\theta = 8^{o}$	$\theta = 46^{o}$	$\theta = 136^{o}$	$\theta = 180^{o}$	$\theta = 8^{o}$	$\theta = 46^{o}$	$\theta = 136^{o}$	$\theta = 180^{o}$
		$\alpha = -5.10^{o}$; $C_L = -0.466$				$\alpha = 1.97^{o}$; $C_L = 0.183$				$\alpha = 7.03^{o}$; $C_L = 0.619$			
153.0	60.22	0.088	0.090	-0.033	0.009	0.063	0.058	-0.007	0.024	0.034	0.025	0.013	0.054
154.4	60.80		.109	-.007	.033		.074	.017	.044		.054	.032	.061
156.1	61.46	.145	.117	.027	.051	.118	.097	.034	.056	.102	.085	.056	.081
157.6	62.04	.166	.124	.048	.073	.138	.110	.057	.069	.129	.099	.074	.092
159.2	62.67		.100	.058	.048		.090	.066	.047		.078	.076	.056
160.8	63.30	.061			.009	.066			.021	.055			.016
		$\alpha = -4.08^{o}$; $C_L = -0.374$				$\alpha = 2.46^{o}$; $C_L = 0.226$				$\alpha \doteq 8.02^{o}$; $C_L = 0.698$			
153.0	60.22	0.080	0.081	-0.033	0.009	0.057	0.052	-0.003	0.031	0.033	0.019	0.017	0.062
154.4	60.80		.098	-.007	.033		.076	.018	.046		.051	.029	.074
156.1	61.46	.134	.114	.023	.047	.116	.095	.041	.055	.102	.073	.055	.080
157.6	62.04	.154	.118	.047	.071	.139	.103	.060	.071	.122	.095	.075	.094
159.2	62.67		.098	.057	.042		.086	.066	.048		.076	.073	.055
160.8	63.30	.061			.011	.060			.021	.053			.013
		$\alpha = -3.04^{o}$; $C_L = -0.279$				$\alpha = 2.99^{o}$; $C_L = 0.272$				$\alpha = 9.04^{o}$; $C_L = 0.776$			
153.0	60.22	0.080	0.075	-0.027	0.013	0.061	0.049	-0.004	0.037	0.033	0.015	0.020	0.061
154.4	60.80		.096	-.006	.036		.075	.020	.053		.045	.034	.076
156.1	61.46	.131	.113	.024	.048	.114	.094	.043	.062	.093	.072	.055	.082
157.6	62.04	.154	.116	.046	.069	.139	.109	.066	.074	.117	.085	.071	.093
159.2	62.67		.092	.057	.047		.086	.073	.050		.069	.077	.056
160.8	63.30	.063			.011	.064			.020	.046			.002
		$\alpha = -2.05^{o}$; $C_L = -0.188$				$\alpha = 3.49^{o}$; $C_L = 0.318$				$\alpha = 10.00^{o}$; $C_L = 0.845$			
153.0	60.22	0.079	0.074	-0.024	0.017	0.054	0.049	-0.001	0.036	0.025	0.003	0.016	0.058
154.4	60.80		.097	.003	.039		.072	.022	.052		.032	.036	.077
156.1	61.46	.134	.109	.033	.052	.116	.095	.048	.062	.086	.063	.053	.078
157.6	62.04	.155	.120	.053	.068	.137	.103	.065	.077	.115	.081	.075	.099
159.2	62.67		.097	.062	.049		.085	.070	.047		.065	.077	.059
160.8	63.30	.065			.021	.060			.019	.048			.006
		$\alpha = -1.08^{o}$; $C_L = -0.097$				$\alpha = 4.00^{o}$; $C_L = 0.366$				$\alpha = 11.02^{o}$; $C_L = 0.913$			
153.0	60.22	0.070	0.065	-0.025	0.016	0.057	0.047	0.000	0.036	0.015	0.002	0.023	0.071
154.4	60.80		.087	.003	.034		.068	.020	.057		.036	.032	.085
156.1	61.46	.125	.102	.028	.050	.112	.090	.048	.059	.086	.060	.058	.084
157.6	62.04	.144	.109	.052	.064	.133	.103	.062	.078	.114	.079	.063	.103
159.2	62.67		.090	.063	.044		.085	.069	.051		.062	.075	.057
160.8	63.30	.063			.020	.060			.019	.049			.003
		$\alpha = -0.02^{o}$; $C_L = 0.003$				$\alpha = 5.01^{o}$; $C_L = 0.450$							
153.0	60.22	0.066	0.062	-0.017	0.025	0.052	0.039	0.006	0.040				
154.4	60.80		.089	.007	.040		.064	.025	.054				
156.1	61.46	.122	.100	.036	.051	.112	.089	.048	.066				
157.6	62.04	.145	.111	.058	.063	.133	.101	.068	.082				
159.2	62.67		.091	.067	.050		.085	.073	.051				
160.8	63.30	.065			.021	.057			.020				
		$\alpha = 1.00^{o}$; $C_L = 0.094$				$\alpha = 6.04^{o}$; $C_L = 0.537$							
153.0	60.22	0.064	0.060	-0.010	0.028	0.042	0.035	0.013	0.049				
154.4	60.80		.085	.012	.044		.061	.025	.064				
156.1	61.46	.118	.100	.042	.058	.109	.086	.048	.071				
157.6	62.04	.140	.108	.056	.067	.128	.096	.071	.085				
159.2	62.67		.088	.071	.049		.077	.073	.055				
160.8	63.30	.067			.026	.058			.008				

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TABLE IX – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5^{\circ}$

(b) M = 0.80

Fuselage station		C_p for –											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -4.96^{\circ}$; $C_L = -0.493$				$\alpha = 1.95^{\circ}$; $C_L = 0.192$				$\alpha = 7.13^{\circ}$; $C_L = 0.685$			
153.0	60.22	0.091	0.086	-0.049	0.000	0.067	0.054	-0.025	0.022	0.041	0.024	0.004	0.049
154.4	60.80		.114	-.014	.032		.090	.011	.046		.066	.033	.064
156.1	61.46	.161	.130	.030	.053	.134	.113	.050	.065	.110	.101	.058	.080
157.6	62.04	.188	.139	.058	.081	.163	.130	.070	.082	.144	.116	.083	.104
159.2	62.67		.113	.074	.055		.106	.088	.060		.094	.091	.066
160.8	63.30	.070			.011	.081			.033	.071			.021
		$\alpha = -4.00^{\circ}$; $C_L = -0.411$				$\alpha = 2.45^{\circ}$; $C_L = 0.240$				$\alpha = 8.20^{\circ}$; $C_L = 0.758$			
153.0	60.22	0.088	0.085	-0.044	-0.003	0.067	0.053	-0.015	0.032	0.032	0.016	0.015	0.053
154.4	60.80		.109	-.010	.031		.085	.020	.051		.058	.038	.070
156.1	61.46	.159	.130	.034	.056	.134	.113	.056	.066	.111	.094	.066	.082
157.6	62.04	.193	.139	.060	.082	.161	.132	.079	.083	.143	.113	.081	.102
159.2	62.67		.116	.077	.055		.110	.087	.062		.096	.086	.065
160.8	63.30	.075			.018	.084			.044	.068			.014
		$\alpha = -2.94^{\circ}$; $C_L = -0.309$				$\alpha = 2.99^{\circ}$; $C_L = 0.290$				$\alpha = 9.11^{\circ}$; $C_L = 0.805$			
153.0	60.22	0.091	0.087	-0.035	0.009	0.064	0.047	-0.013	0.031	0.029	0.010	0.019	0.064
154.4	60.80		.111	.001	.032		.077	.022	.053		.055	.035	.085
156.1	61.46	.156	.129	.042	.058	.132	.115	.052	.063	.110	.093	.064	.092
157.6	62.04	.184	.146	.067	.086	.163	.130	.080	.088	.139	.119	.085	.117
159.2	62.67		.115	.083	.061		.105	.084	.066		.099	.085	.069
160.8	63.30	.082			.028	.084			.034	.065			.021
		$\alpha = -2.01^{\circ}$; $C_L = -0.214$				$\alpha = 3.52^{\circ}$; $C_L = 0.341$				$\alpha = 10.09^{\circ}$; $C_L = 0.861$			
153.0	60.22	0.082	0.076	-0.035	0.002	0.061	0.049	-0.006	0.031	0.015	-0.004	0.022	0.064
154.4	60.80		.104	-.002	.031		.084	.023	.056		.039	.038	.082
156.1	61.46	.155	.121	.037	.055	.128	.112	.057	.067	.100	.084	.060	.083
157.6	62.04	.181	.138	.068	.081	.155	.131	.077	.088	.129	.104	.076	.106
159.2	62.67		.112	.076	.062		.106	.087	.062		.090	.085	.071
160.8	63.30	.078			.032	.180			.034	.057			.011
		$\alpha = -1.05^{\circ}$; $C_L = -0.119$				$\alpha = 4.04^{\circ}$; $C_L = 0.393$				$\alpha = 11.18^{\circ}$; $C_L = 0.933$			
153.0	60.22	0.080	0.066	-0.029	0.009	0.055	0.041	-0.012	0.034	0.010	-0.000	0.013	0.068
154.4	60.80		.100	.006	.039		.078	.021	.046		.042	.040	.086
156.1	61.46	.142	.124	.045	.058	.127	.109	.055	.068	.095	.085	.059	.098
157.6	62.04	.172	.138	.068	.082	.154	.126	.078	.091	.130	.106	.086	.121
159.2	62.67		.110	.077	.064		.104	.086	.069		.097	.079	.065
160.8	63.30	.076			.037	.079			.033	.061			.013
		$\alpha = 0.02^{\circ}$; $C_L = -0.007$				$\alpha = 5.03^{\circ}$; $C_L = 0.495$							
153.0	60.22	0.073	0.063	-0.030	0.013	0.057	0.036	0.001	0.036				
154.4	60.80		.093	.005	.036		.076	.028	.064				
156.1	61.46	.145	.116	.039	.056	.121	.106	.058	.073				
157.6	62.04	.165	.131	.067	.074	.151	.123	.083	.099				
159.2	62.67		.109	.078	.059		.109	.093	.068				
160.8	63.30	.078			.032	.079			.026				
		$\alpha = 1.03^{\circ}$; $C_L = 0.096$				$\alpha = 6.05^{\circ}$; $C_L = 0.591$							
153.0	60.22	0.069	0.058	-0.017	0.018	0.046	0.028	-0.003	0.046				
154.4	60.80		.096	.014	.042		.068	.025	.071				
156.1	61.46	.144	.116	.048	.057	.120	.097	.057	.080				
157.6	62.04	.168	.133	.073	.084	.147	.120	.072	.099				
159.2	62.67		.116	.081	.063		.104	.091	.066				
160.8	63.30	.081			.042	.078			.027				

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TABLE IX – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5^o$

(c) M = 0.90

Fuselage station		C_p for –											
cm	in.	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$
		$\alpha = -4.87^o$; $C_L = -0.520$				$\alpha = 1.03^o$; $C_L = 0.094$				$\alpha = 5.03^o$; $C_L = 0.553$			
153.0	60.22	0.091	0.077	-0.054	-0.002	0.072	0.057	-0.030	0.011	0.053	0.030	-0.010	0.033
154.4	60.80		.108	-.007	.035		.099	.013	.040		.080	.028	.062
156.1	61.46	.169	.140	.039	.064	.153	.131	.056	.065	.137	.117	.065	.078
157.6	62.04	.204	.151	.068	.099	.185	.152	.082	.090	.166	.142	.090	.104
159.2	62.67		.126	.088	.078		.129	.099	.079		.119	.101	.080
160.8	63.30	.081			.026	.101			.054	.097			.046
		$\alpha = -3.96^o$; $C_L = -0.441$				$\alpha = 1.95^o$; $C_L = 0.199$				$\alpha = 6.10^o$; $C_L = 0.674$			
153.0	60.22	0.089	0.073	-0.045	-0.008	0.066	0.052	-0.019	0.017	0.046	0.026	-0.008	0.036
154.4	60.80		.107	-.002	.029		.092	.014	.046		.073	.030	.066
156.1	61.46	.165	.140	.043	.062	.150	.124	.059	.070	.128	.117	.064	.081
157.6	62.04	.199	.151	.069	.089	.179	.149	.084	.091	.162	.140	.092	.109
159.2	62.67		.129	.086	.072		.126	.096	.075		.123	.100	.076
160.8	63.30	.087			.032	.100			.050	.089			.034
		$\alpha = -2.95^o$; $C_L = -0.344$				$\alpha = 2.45^o$; $C_L = 0.250$				$\alpha = 7.17^o$; $C_L = 0.771$			
153.0	60.22	0.084	0.068	-0.048	-0.004	0.066	0.045	-0.023	0.018	0.042	0.015	0.002	0.039
154.4	60.80		.103	-.003	.032		.091	.020	.046		.067	.030	.068
156.1	61.46	.163	.136	.042	.061	.145	.127	.055	.075	.130	.114	.066	.085
157.6	62.04	.195	.151	.073	.093	.176	.154	.088	.090	.156	.133	.089	.111
159.2	62.67		.124	.088	.074		.127	.099	.078		.120	.100	.077
160.8	63.30	.089			.037	.099			.051	.088			.035
		$\alpha = -1.98^o$; $C_L = -0.239$				$\alpha = 3.01^o$; $C_L = 0.312$				$\alpha = 7.90^o$; $C_L = 0.821$			
153.0	60.22	0.088	0.068	-0.040	-0.005	0.064	0.047	-0.018	0.017	0.036	0.011	0.001	0.052
154.4	60.80		.106	.001	.034		.090	.019	.050		.063	.037	.076
156.1	61.46	.165	.135	.045	.060	.146	.129	.063	.070	.126	.111	.067	.087
157.6	62.04	.195	.151	.070	.090	.174	.149	.084	.097	.156	.133	.093	.116
159.2	62.67		.128	.092	.077		.126	.098	.080		.117	.103	.080
160.8	63.30	.091			.047	.098			.049	.086			.035
		$\alpha = -1.02^o$; $C_L = -0.132$				$\alpha = 3.50^o$; $C_L = 0.367$							
153.0	60.22	0.083	0.063	-0.040	-0.003	0.062	0.040	-0.019	0.022				
154.4	60.80		.103	.003	.033		.083	.020	.052				
156.1	61.46	.162	.134	.046	.063	.144	.124	.060	.073				
157.6	62.04	.190	.149	.075	.088	.171	.148	.086	.095				
159.2	62.67		.127	.089	.078		.126	.099	.076				
160.8	63.30	.094			.057	.099			.054				
		$\alpha = 0.03^o$; $C_L = -0.017$				$\alpha = 4.01^o$; $C_L = 0.430$							
153.0	60.22	0.074	0.061	-0.032	0.009	0.056	0.037	-0.019	0.025				
154.4	60.80		.103	.008	.035		.083	.023	.054				
156.1	61.46	.157	.133	.051	.065	.140	.118	.062	.072				
157.6	62.04	.184	.155	.080	.091	.166	.142	.086	.097				
159.2	62.67		.127	.091	.075		.122	.102	.074				
160.8	63.30	.098			.054	.097			.045				

[redacted]

TABLE IX – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5^{\circ}$

(d) M = 0.95

Fuselage station		C_p for –											
cm	in.	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -4.82^{\circ}$; $C_L = -0.547$				$\alpha = 1.04^{\circ}$; $C_L = 0.097$				$\alpha = 4.01^{\circ}$; $C_L = 0.456$			
153.0	60.22	0.088	0.069	-0.092	-0.035	0.068	0.052	-0.041	-0.005	0.056	0.033	-0.020	0.022
154.4	60.80		.112	-.012	.023		.096	.017	.037		.085	.024	.054
156.1	61.46	.179	.143	.042	.068	.159	.137	.064	.070	.149	.130	.071	.085
157.6	62.04	.216	.158	.075	.110	.194	.167	.097	.098	.182	.160	.100	.110
159.2	62.67		.137	.098	.089		.145	.114	.088		.138	.112	.093
160.8	63.30	.089			.046	.115			.075	.112			.064
		$\alpha = -3.91^{\circ}$; $C_L = -0.470$				$\alpha = 1.97^{\circ}$; $C_L = 0.211$				$\alpha = 5.04^{\circ}$; $C_L = 0.571$			
153.0	60.22	0.086	0.067	-0.090	-0.043	0.067	0.049	-0.033	0.008	0.046	0.030	-0.013	0.029
154.4	60.80		.107	-.015	.021		.099	.024	.047		.086	.034	.066
156.1	61.46	.177	.141	.043	.061	.160	.140	.071	.074	.145	.131	.076	.089
157.6	62.04	.216	.161	.077	.100	.189	.166	.102	.106	.174	.160	.104	.118
159.2	62.67		.136	.099	.085		.151	.118	.093		.141	.115	.097
160.8	63.30	.094			.050	.116			.073	.112			.062
		$\alpha = -2.96^{\circ}$; $C_L = -0.377$				$\alpha = 2.20^{\circ}$; $C_L = 0.246$				$\alpha = 6.02^{\circ}$; $C_L = 0.670$			
153.0	60.22	0.081	0.067	-0.079	-0.039	0.065	0.046	-0.029	0.007	0.043	0.024	-0.006	0.037
154.4	60.80		.107	-.008	.023		.096	.022	.046		.082	.039	.071
156.1	61.46	.175	.143	.048	.066	.156	.140	.067	.078	.145	.130	.079	.090
157.6	62.04	.209	.164	.086	.101	.187	.164	.103	.100	.172	.156	.103	.122
159.2	62.67		.140	.108	.092		.146	.117	.092		.143	.117	.092
160.8	63.30	.105			.057	.117			.068	.111			.054
		$\alpha = -1.98^{\circ}$; $C_L = -0.264$				$\alpha = 2.48^{\circ}$; $C_L = 0.270$				$\alpha = 7.10^{\circ}$; $C_L = 0.767$			
153.0	60.22	0.083	0.058	-0.069	-0.026	0.068	0.047	-0.031	0.010	0.040	0.016	-0.001	0.047
154.4	60.80		.104	-.002	.029		.095	.020	.047		.078	.036	.074
156.1	61.46	.171	.144	.051	.068	.155	.137	.070	.079	.138	.129	.078	.099
157.6	62.04	.206	.160	.088	.098	.186	.166	.099	.105	.170	.152	.104	.124
159.2	62.67		.143	.105	.090		.145	.113	.095		.137	.116	.093
160.8	63.30	.104			.064	.119			.077	.109			.051
		$\alpha = -1.07^{\circ}$; $C_L = -0.154$				$\alpha = 3.03^{\circ}$; $C_L = 0.335$				$\alpha = 8.22^{\circ}$; $C_L = 0.856$			
153.0	60.22	0.078	0.064	-0.058	-0.021	0.060	0.042	-0.028	0.011	0.032	0.007	0.003	0.046
154.4	60.80		.105	.007	.029		.091	.022	.047		.070	.039	.076
156.1	61.46	.167	.139	.057	.069	.152	.136	.069	.079	.136	.117	.075	.094
157.6	62.04	.202	.166	.090	.097	.181	.160	.099	.101	.163	.146	.102	.120
159.2	62.67		.145	.107	.092		.144	.115	.090		.127	.107	.091
160.8	63.30	.109			.072	.111			.067	.100			.046
		$\alpha = 0.03^{\circ}$; $C_L = -0.021$				$\alpha = 3.53^{\circ}$; $C_L = 0.396$				$\alpha = 9.22^{\circ}$; $C_L = 0.928$			
153.0	60.22	0.073	0.059	-0.050	-0.014	0.063	0.042	-0.021	0.019	0.031	0.009	0.011	0.058
154.4	60.80		.102	.008	.031		.094	.029	.055		.070	.046	.088
156.1	61.46	.160	.141	.059	.069	.153	.134	.072	.080	.134	.119	.083	.101
157.6	62.04	.191	.163	.089	.098	.183	.162	.104	.113	.167	.147	.105	.129
159.2	62.67		.142	.110	.088		.142	.116	.095		.129	.115	.098
160.8	63.30	.111			.069	.121			.073	.106			.044

TABLE IX – Continued

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5^{\circ}$

(e) M = 0.97

Fuselage station, cm	Fuselage station, in.	C_p for – $\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$	$\theta = 8^{\circ}$	$\theta = 46^{\circ}$	$\theta = 136^{\circ}$	$\theta = 180^{\circ}$
		$\alpha = -4.94^{\circ}$; $C_L = -0.558$				$\alpha = 1.89^{\circ}$; $C_L = 0.200$				$\alpha = 7.03^{\circ}$; $C_L = 0.765$			
153.0	60.22	0.099	0.075	-0.151	-0.102	0.066	0.044	-0.077	-0.030	0.042	0.018	0.000	0.049
154.4	60.80		.112	-.042	-.009		.096	.003	.029		.083	.038	.080
156.1	61.46	.193	.145	.032	.054	.167	.142	.069	.073	.143	.132	.082	.103
157.6	62.04	.237	.167	.077	.106	.199	.170	.105	.106	.173	.159	.112	.132
159.2	62.67		.141	.103	.098		.155	.122	.101		.146	.122	.106
160.8	63.30	.100			.059	.129			.086	.114			.061
		$\alpha = -3.93^{\circ}$; $C_L = -0.477$				$\alpha = 2.39^{\circ}$; $C_L = 0.257$				$\alpha = 8.09^{\circ}$; $C_L = 0.853$			
153.0	60.22	0.087	0.066	-0.145	-0.110	0.059	0.040	-0.073	-0.025	0.039	0.018	0.007	0.055
154.4	60.80		.105	-.046	-.020		.095	.005	.039		.084	.045	.089
156.1	61.46	.186	.141	.032	.046	.160	.143	.068	.078	.140	.134	.089	.109
157.6	62.04	.228	.164	.076	.097	.195	.172	.103	.108	.173	.164	.115	.139
159.2	62.67		.140	.100	.091		.152	.123	.099		.146	.125	.108
160.8	63.30	.099			.057	.125			.079	.119			.058
		$\alpha = -2.94^{\circ}$; $C_L = -0.379$				$\alpha = 2.93^{\circ}$; $C_L = 0.324$				$\alpha = 9.19^{\circ}$; $C_L = 0.934$			
153.0	60.22	0.079	0.056	-0.134	-0.104	0.062	0.035	-0.053	-0.006	0.029	0.011	0.013	0.064
154.4	60.80		.105	-.042	-.019		.095	.019	.042		.079	.050	.093
156.1	61.46	.177	.139	.034	.047	.160	.139	.073	.080	.141	.133	.086	.108
157.6	62.04	.214	.160	.077	.094	.191	.170	.108	.111	.172	.158	.114	.138
159.2	62.67		.139	.102	.090		.152	.123	.102		.143	.122	.109
160.8	63.30	.104			.064	.123			.080	.121			.056
		$\alpha = -1.98^{\circ}$; $C_L = -0.266$				$\alpha = 3.46^{\circ}$; $C_L = 0.388$				$\alpha = 10.13^{\circ}$; $C_L = 0.998$			
153.0	60.22	0.073	0.053	-0.121	-0.085	0.057	0.037	-0.042	0.005	0.021	0.002	0.015	0.069
154.4	60.80		.100	-.030	-.006		.092	.022	.044		.078	.045	.095
156.1	61.46	.178	.141	.041	.051	.155	.140	.078	.083	.138	.125	.084	.111
157.6	62.04	.213	.168	.086	.099	.187	.166	.106	.116	.170	.155	.115	.144
159.2	62.67		.145	.107	.094		.151	.123	.102		.143	.123	.111
160.8	63.30	.116			.076	.124			.078	.118			.052
		$\alpha = -1.08^{\circ}$; $C_L = -0.156$				$\alpha = 3.96^{\circ}$; $C_L = 0.449$				$\alpha = 11.17^{\circ}$; $C_L = 1.052$			
153.0	60.22	0.069	0.045	-0.114	-0.075	0.058	0.034	-0.034	0.020	0.018	0.002	0.029	0.078
154.4	60.80		.096	-.029	-.002		.095	.031	.059		.074	.056	.106
156.1	61.46	.170	.137	.048	.056	.155	.139	.081	.091	.131	.125	.089	.121
157.6	62.04	.207	.165	.085	.096	.188	.169	.111	.120	.166	.157	.116	.147
159.2	62.67		.148	.113	.095		.151	.128	.105		.142	.123	.113
160.8	63.30	.116			.077	.124			.078	.124			.059
		$\alpha = -0.03^{\circ}$; $C_L = -0.029$				$\alpha = 4.95^{\circ}$; $C_L = 0.560$							
153.0	60.22	0.071	0.047	-0.094	-0.053	0.051	0.032	-0.015	0.027				
154.4	60.80		.099	-.015	.014		.089	.039	.065				
156.1	61.46	.170	.141	.055	.060	.151	.139	.083	.092				
157.6	62.04	.205	.166	.091	.100	.184	.167	.112	.125				
159.2	62.67		.152	.115	.097		.152	.127	.103				
160.8	63.30	.122			.081	.121			.074				
		$\alpha = 0.98^{\circ}$; $C_L = 0.092$				$\alpha = 5.98^{\circ}$; $C_L = 0.665$							
153.0	60.22	0.068	0.044	-0.099	-0.046	0.043	0.024	-0.011	0.035				
154.4	60.80		.096	-.007	.018		.086	.040	.073				
156.1	61.46	.167	.139	.058	.067	.145	.133	.084	.097				
157.6	62.04	.202	.168	.100	.101	.177	.164	.114	.127				
159.2	62.67		.152	.118	.098		.151	.125	.101				
160.8	63.30	.125			.084	.121			.069				

TABLE IX – Concluded

PRESSURE DISTRIBUTIONS OVER REAR FUSELAGE WITH FUSELAGE AREA-RULE ADDITIONS; $\delta_h = -5^o$

(f) M = 0.99

Fuselage station		C_p for –											
cm	in.	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$	$\theta = 8^o$	$\theta = 46^o$	$\theta = 136^o$	$\theta = 180^o$
		$\alpha = -4.94^o$; $C_L = -0.569$				$\alpha = 1.89^o$; $C_L = 0.209$				$\alpha = 6.99^o$; $C_L = 0.760$			
153.0	60.22	0.113	0.068	-0.194	-0.160	0.060	0.026	-0.117	-0.067	0.055	0.036	0.016	0.064
154.4	60.80		.113	-.090	-.068		.089	-.039	-.012		.101	.070	.098
156.1	61.46	.206	.149	.008	.017	.170	.140	.044	.044	.162	.151	.112	.130
157.6	62.04	.255	.166	.065	.082	.211	.172	.099	.093	.197	.181	.134	.156
159.2	62.67		.138	.102	.089		.162	.127	.098		.169	.149	.133
160.8	63.30	.102			.064	.133			.090	.141			.089
		$\alpha = -3.94^o$; $C_L = -0.482$				$\alpha = 2.38^o$; $C_L = 0.267$				$\alpha = 8.04^o$; $C_L = 0.850$			
153.0	60.22	0.103	0.068	-0.188	-0.148	0.047	0.016	-0.118	-0.070	0.047	0.027	0.017	0.059
154.4	60.80		.103	-.088	-.065		.083	-.042	-.015		.096	.063	.099
156.1	61.46	.201	.142	.015	.013	.164	.135	.047	.049	.158	.150	.105	.128
157.6	62.04	.248	.163	.067	.080	.202	.170	.097	.095	.193	.179	.133	.157
159.2	62.67		.142	.096	.085		.156	.123	.099		.163	.146	.132
160.8	63.30	.100			.064	.129			.092	.140			.086
		$\alpha = -2.94^o$; $C_L = -0.382$				$\alpha = 2.92^o$; $C_L = 0.333$				$\alpha = 9.16^o$; $C_L = 0.934$			
153.0	60.22	0.095	0.059	-0.153	-0.121	0.051	0.024	-0.092	-0.038	0.038	0.022	0.012	0.063
154.4	60.80		.103	-.070	-.053		.093	-.008	.010		.094	.060	.101
156.1	61.46	.194	.149	.018	.021	.168	.144	.060	.065	.156	.147	.102	.127
157.6	62.04	.237	.166	.071	.083	.207	.178	.104	.108	.184	.173	.126	.156
159.2	62.67		.147	.106	.089		.158	.130	.108		.161	.140	.129
160.8	63.30	.109			.070	.133			.092	.141			.080
		$\alpha = -1.97^o$; $C_L = -0.282$				$\alpha = 3.44^o$; $C_L = 0.397$				$\alpha = 10.11^o$; $C_L = 1.004$			
153.0	60.22	0.081	0.050	-0.154	-0.104	0.058	0.030	-0.068	-0.023	0.023	0.009	0.018	0.067
154.4	60.80		.101	-.064	-.042		.092	.012	.035		.089	.062	.100
156.1	61.46	.188	.144	.022	.021	.165	.146	.069	.077	.148	.140	.095	.123
157.6	62.04	.234	.168	.074	.083	.205	.175	.110	.116	.183	.170	.127	.158
159.2	62.67		.151	.111	.089		.161	.132	.109		.160	.136	.128
160.8	63.30	.113			.080	.129			.086	.139			.076
		$\alpha = -1.06^o$; $C_L = -0.172$				$\alpha = 3.94^o$; $C_L = 0.455$				$\alpha = 11.15^o$; $C_L = 1.060$			
153.0	60.22	0.081	0.046	-0.122	-0.082	0.064	0.043	-0.030	0.013	0.017	0.019	0.024	0.080
154.4	60.80		.100	-.049	-.021		.104	.039	.063		.087	.061	.114
156.1	61.46	.184	.147	.037	.035	.176	.155	.091	.101	.151	.142	.096	.136
157.6	62.04	.231	.172	.082	.087	.212	.183	.123	.126	.184	.174	.126	.162
159.2	62.67		.152	.116	.098		.168	.143	.119		.160	.142	.133
160.8	63.30	.124			.091	.139			.095	.143			.078
		$\alpha = -0.05^o$; $C_L = -0.041$				$\alpha = 4.96^o$; $C_L = 0.566$							
153.0	60.22	0.075	0.042	-0.100	-0.053	0.062	0.038	-0.009	0.044				
154.4	60.80		.100	-.023	-.004		.105	.055	.078				
156.1	61.46	.186	.145	.047	.058	.170	.155	.101	.113				
157.6	62.04	.226	.178	.097	.097	.205	.182	.127	.141				
159.2	62.67		.160	.123	.101		.172	.145	.125				
160.8	63.30	.130			.097	.137			.089				
		$\alpha = 0.97^o$; $C_L = 0.088$				$\alpha = 6.00^o$; $C_L = 0.667$							
153.0	60.22	0.069	0.034	-0.111	-0.059	0.060	0.040	0.011	0.052				
154.4	60.80		.095	-.026	-.003		.105	.064	.096				
156.1	61.46	.180	.143	.052	.053	.164	.155	.107	.123				
157.6	62.04	.220	.178	.098	.100	.203	.186	.135	.153				
159.2	62.67		.160	.126	.102		.173	.147	.131				
160.8	63.30	.130			.095	.137			.090				

[redacted]

TABLE X

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(a) $M = 0.50$; $\beta = -5.3^{\circ}$.

	Orifice number	C_p at α, deg, of —										
		-6.58	-4.37	-2.17	0.09	2.48	4.45	6.72	8.93	11.29	13.36	15.20
Fore fuselage area-rule addition	101	.2530	.2952	.2257	.1871	.1674	.1465	.1110	.0875	.0471	.0324	-.0159
	102	.2022	.1724	.1421	.1022	.0625	.0424	-.0037	-.0266	-.0617	-.0875	-.1079
	103	-.0448	-.0225	-.0115	.0001	.0011	.0033	-.0087	-.0259	-.0360	-.0567	-.0844
	104	-.0724	-.0231	.0111	.0299	.0440	.0647	.0753	.0879	.0759	.0597	.0528
	105	.0274	.0136	.0026	-.0137	-.0331	-.0470	-.0737	-.0960	-.1336	-.1557	-.1802
	106	-.0947	-.0664	-.0429	-.0407	-.0319	-.0294	-.0372	-.0514	-.0809	-.1004	-.1384
	107	-.1880	-.1201	-.0708	-.0218	.0130	.0380	.0553	.0669	.0671	.0732	.0534
	108	-.0159	-.0394	-.0608	-.0755	-.0975	-.1120	-.1399	-.1591	-.1848	-.2063	-.2282
	109	-.1305	-.1038	-.0947	-.0940	-.0956	-.0922	-.1076	-.1321	-.1571	-.1796	-.2147
	110	-.1582	-.1333	-.1139	-.0972	-.0959	-.0919	-.1022	-.1186	-.1487	-.1730	-.1952
	111	-.1704	-.1305	-.0991	-.0774	-.0642	-.0489	-.0489	-.0589	-.0752	-.0970	-.1155
	112	-.2172	-.1798	-.1324	-.0897	-.0561	-.0256	-.0099	.0089	.0195	.0127	.0205
	113	-.0043	-.0033	-.0102	-.0137	-.0153	-.0184	-.0256	-.0247	-.0382	-.0385	-.0388
	114	-.0994	-.0843	-.0771	-.0799	-.0755	-.0784	-.0963	-.1101	-.1302	-.1529	-.1676
	115	-.1848	-.1421	-.1255	-.1141	-.0969	-.0913	-.1063	-.1148	-.1418	-.1677	-.1978
	116	-.2210	-.1854	-.1538	-.1267	-.1085	-.0995	-.0988	-.1007	-.1123	-.1328	-.1560
	117	-.1092	-.0705	-.0536	-.0206	.0101	.0418	.0606	.0716	.1025	.1193	.1302
	118	-.1607	-.1198	-.1048	-.0714	-.0589	-.0388	-.0394	-.0482	-.0476	-.0615	-.0539
	119	-.1855	-.1625	-.1252	-.0966	-.0762	-.0649	-.0661	-.0664	-.0834	-.0929	-.1167
	120	-.1619	-.1295	-.1026	-.0784	-.0649	-.0533	-.0586	-.0671	-.0793	-.0976	-.1114
	121	-.1638	-.1277	-.1029	-.0802	-.0564	-.0454	-.0372	-.0369	-.0451	-.0495	-.0671
	122	-.1456	-.0884	-.0545	-.0168	.0377	.0782	.1070	.1392	.1752	.2020	.2160
	123	-.1274	-.0708	-.0344	.0083	.0424	.0816	.1135	.1418	.1749	.2149	.2289
	124	-.1095	-.0715	-.0369	.0004	.0421	.0656	.0935	.1157	.1386	.1679	.1957
	125	-.0881	-.0488	-.0124	.0186	.0393	.0628	.0756	.0957	.1144	.1177	.1346
	126	-.0859	-.0394	-.0143	.0148	.0421	.0619	.0731	.0791	.0841	.0870	.0913
Aft fuselage area-rule addition	127	.0908	.1266	.1196	.1192	.1345	.1491	.1623	.1677	.1638	.1463	.1355
	128	.0496	.0686	.0828	.0986	.1155	.1329	.1481	.1617	.1727	.1726	.1665
	129	.0600	.0591	.0584	.0568	.0521	.0569	.0657	.0736	.0619	.0572	.0281
	130	-.1358	-.1072	-.0749	-.0435	-.0187	-.0011	.0264	.0420	.0524	.0521	.0598
	131	-.0071	-.0061	-.0099	-.0096	-.0159	-.0074	.0008	.0001	-.0046	-.0197	-.0489
	132	-.0341	-.0306	-.0288	-.0262	-.0181	-.0159	-.0102	-.0058	-.0140	-.0360	-.0686
	133	-.0630	-.0479	-.0407	-.0363	-.0300	-.0225	-.0184	-.0190	-.0379	-.0536	-.0708
	134	-.0865	-.0749	-.0655	-.0510	-.0423	-.0259	-.0121	-.0027	-.0131	-.0159	-.0467
	135	-.1226	-.1028	-.0796	-.0639	-.0470	-.0300	-.0099	-.0058	-.0068	-.0093	-.0278
	136	-.1540	-.1295	-.1010	-.0771	-.0551	-.0363	-.0143	.0001	.0096	.0043	-.0030
	137	-.0586	-.0523	-.0539	-.0667	-.0686	-.0614	-.0542	-.0558	-.0664	-.0756	-.0856
	138	-.1044	-.0962	-.0869	-.0824	-.0802	-.0655	-.0655	-.0677	-.0718	-.0982	-.1092
	139	-.1226	-.1138	-.0969	-.0871	-.0758	-.0727	-.0639	-.0702	-.0793	-.0680	-.0646
	140	-.1374	-.1185	-.1026	-.0928	-.0812	-.0668	-.0589	-.0507	-.0463	-.0552	-.0508
	141	-.1657	-.1380	-.1173	-.1044	-.0859	-.0702	-.0517	-.0473	-.0444	-.0360	-.0457
	142	-.1707	-.1559	-.1346	-.1088	-.0871	-.0690	-.0539	-.0429	-.0309	-.0385	-.0573
	143	.0071	.0090	-.0011	-.0131	-.0322	-.0404	-.0454	-.0611	-.0655	-.0872	-.1076
	144	-.0253	-.0272	-.0332	-.0448	-.0423	-.0335	-.0363	-.0438	-.0608	-.0702	-.0796
	145	-.0639	-.0601	-.0441	-.0372	-.0275	-.0197	-.0234	-.0363	-.0259	-.0285	-.0558
	146	-.1063	-.0834	-.0649	-.0507	-.0316	-.0269	-.0203	-.0046	-.0124	-.0209	-.0344
	147	-.1902	-.1609	-.1368	-.0981	-.0774	-.0608	-.0291	-.0234	-.0153	-.0027	-.0165

[redacted]

TABLE X - Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(b) M = 0.80; β = -5.8°.

	Orifice number	C_p at α, deg, of —							
		-7.40	-4.96	-2.28	0.29	2.91	5.49	7.95	10.29
Fore fuselage area-rule addition	101	.3159	.2952	.2647	.2184	.1920	.1727	.1355	.0968
	102	.2254	.1938	.1587	.1109	.0764	.0406	.0164	-.0168
	103	-.0454	-.0217	.0012	.0001	.0038	.0074	.0052	-.0048
	104	-.0820	-.0138	.0211	.0389	.0616	.0824	.1081	.1147
	105	.0263	.0125	-.0086	-.0330	-.0464	-.0659	-.0912	-.1155
	106	-.1108	-.0815	-.0560	-.0464	-.0374	-.0309	-.0369	-.0550
	107	-.2251	-.1533	-.0818	-.0253	.0140	.0568	.0871	.0915
	108	-.0215	-.0543	-.0816	-.1073	-.1318	-.1565	-.1788	-.2046
	109	-.1481	-.1247	-.1130	-.1136	-.1148	-.1155	-.1349	-.1580
	110	-.1822	-.1505	-.1284	-.1150	-.1082	-.1122	-.1212	-.1374
	111	-.1974	-.1500	-.1122	-.0933	-.0722	-.0599	-.0582	-.0613
	112	-.2497	-.2020	-.1468	-.0996	-.0624	-.0254	.0049	.0195
	113	.0137	.0026	-.0065	-.0080	-.0122	-.0073	-.0045	-.0005
	114	-.1076	-.0913	-.0815	-.0837	-.0854	-.0886	-.0944	-.1128
	115	-.1981	-.1672	-.1448	-.1062	-.1067	-.1032	-.1139	-.1326
	116	-.2649	-.2125	-.1759	-.1424	-.1235	-.1072	-.1070	-.1144
	117	-.1271	-.0914	-.0518	-.0226	.0142	.0594	.0992	.1249
	118	-.1916	-.1516	-.1153	-.0813	-.0608	-.0412	-.0232	-.0268
	119	-.2247	-.1790	-.1363	-.1104	-.0892	-.0755	-.0686	-.0711
	120	-.1890	-.1456	-.1166	-.0928	-.0745	-.0594	-.0572	-.0681
	121	-.1863	-.1544	-.1158	-.0982	-.0681	-.0487	-.0405	-.0360
	122	-.1389	-.1020	-.0560	-.0120	.0445	.0973	.1484	.1944
	123	-.1282	-.0780	-.0380	.0114	.0568	.1019	.1499	.1975
	124	-.1119	-.0717	-.0287	.0096	.0459	.0882	.1238	.1697
	125	-.0914	-.0472	-.0114	.0187	.0502	.0849	.1158	.1432
	126	-.0808	-.0528	-.0058	.0220	.0533	.0799	.1113	.1163
Aft fuselage area-rule addition	127	.1170	.1266	.1426	.1516	.1685	.1930	.1804	.1611
	128	.0717	.0836	.1048	.1304	.1481	.1660	.1788	.1705
	129	.0569	.0699	.0667	.0646	.0699	.0735	.0494	.0271
	130	-.1396	-.1135	-.0788	-.0490	-.0109	.0106	.0192	.0314
	131	-.0149	-.0097	-.0180	-.0199	-.0029	.0017	-.0283	-.0223
	132	-.0551	-.0355	-.0320	-.0281	-.0172	-.0068	-.0261	-.0259
	133	-.0757	-.0648	-.0519	-.0383	-.0267	-.0103	-.0401	-.0442
	134	-.1056	-.0950	-.0761	-.0600	-.0366	-.0212	-.0341	-.0272
	135	-.1347	-.1147	-.0958	-.0703	-.0413	-.0194	-.0325	-.0251
	136	-.1675	-.1472	-.1159	-.0844	-.0544	-.0319	-.0240	-.0223
	137	-.0696	-.0629	-.0709	-.0770	-.0741	-.0687	-.0761	-.0426
	138	-.1284	-.1196	-.1098	-.1007	-.0852	-.0791	-.0812	-.0605
	139	-.1421	-.1341	-.1186	-.1033	-.0882	-.0739	-.0818	-.0731
	140	-.1561	-.1472	-.1273	-.1071	-.0911	-.0761	-.0739	-.0429
	141	-.1757	-.1605	-.1399	-.1195	-.1019	-.0722	-.0681	-.0549
	142	-.1922	-.1761	-.1547	-.1307	-.1003	-.0780	-.0646	-.0525
	143	.0090	.0061	-.0060	-.0161	-.0383	-.0440	-.0475	-.0514
	144	-.0182	-.0262	-.0375	-.0457	-.0423	-.0398	-.0431	-.0479
	145	-.0391	-.0472	-.0438	-.0314	-.0309	-.0270	-.0416	-.0303
	146	-.0535	-.0557	-.0416	-.0412	-.0249	-.0235	-.0328	-.0105
	147	-.1216	-.1207	-.1044	-.0944	-.0665	-.0445	-.0371	-.0279

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(c) M = 0.90; β = -5.9°.

	Orifice number	C_p at α, deg, of —							
		-7.59	-5.00	-2.39	0.26	2.92	5.59	8.11	9.93
Fore fuselage area-rule addition	101	.3470	.3063	.2843	.2432	.2115	.1866	.1562	.1333
	102	.2532	.2177	.1683	.1282	.0868	.0644	.0358	.0132
	103	-.0324	-.0124	.0009	.0011	.0148	.0311	.0184	.0141
	104	-.0632	-.0145	.0188	.0560	.0664	.1012	.1214	.1391
	105	.0162	.0019	-.0176	-.0392	-.0530	-.0685	-.0854	-.0985
	106	-.1222	-.0865	-.0675	-.0532	-.0415	-.0338	-.0373	-.0410
	107	-.2296	-.1592	-.0895	-.0269	.0124	.0605	.0876	.1096
	108	-.0272	-.0676	-.0994	-.1304	-.1624	-.1869	-.2146	-.2343
	109	-.1682	-.1406	-.1347	-.1334	-.1399	-.1413	-.1557	-.1743
	110	-.1960	-.1649	-.1450	-.1307	-.1312	-.1283	-.1355	-.1411
	111	-.2078	-.1592	-.1230	-.1019	-.0826	-.0677	-.0657	-.0642
	112	-.2560	-.2026	-.1509	-.1032	-.0609	-.0283	-.0044	.0290
	113	.0135	.0030	-.0016	.0008	.0019	.0151	.0232	.0380
	114	-.1026	-.0857	-.0845	-.0832	-.0785	-.0827	-.0903	-.0963
	115	-.1976	-.1671	-.1471	-.1228	-.1116	-.1074	-.1188	-.1300
	116	-.2711	-.2163	-.1817	-.1545	-.1339	-.1194	-.1117	-.1119
	117	-.1340	-.0970	-.0753	-.0382	.0170	.0698	.1143	.1382
	118	-.2088	-.1628	-.1329	-.0925	-.0659	-.0390	-.0107	-.0060
	119	-.2421	-.2006	-.1597	-.1295	-.0946	-.0704	-.0565	-.0536
	120	-.2003	-.1608	-.1331	-.1079	-.0865	-.0683	-.0634	-.0540
	121	-.1943	-.1590	-.1343	-.1004	-.0790	-.0538	-.0362	-.0310
	122	-.1325	-.0936	-.0511	-.0129	.0421	.1078	.1639	.1994
	123	-.1104	-.0700	-.0285	.0099	.0543	.1155	.1597	.2077
	124	-.1002	-.0690	-.0281	.0108	.0479	.1044	.1454	.1890
	125	-.0744	-.0412	-.0102	.0250	.0553	.0973	.1333	.1642
	126	-.0676	-.0329	-.0006	.0298	.0568	.0894	.1232	.1444
Aft fuselage area-rule addition	127	.1380	.1420	.1525	.1726	.1825	.1955	.2026	.1926
	128	.0994	.1056	.1175	.1369	.1625	.1844	.2040	.1897
	129	.0751	.0837	.0783	.0805	.0820	.0910	.0807	.0069
	130	-.1350	-.1084	-.0801	-.0410	-.0104	.0220	.0328	.0150
	131	-.0245	-.0176	-.0135	-.0116	-.0059	.0074	-.0201	-.0747
	132	-.0731	-.0500	-.0365	-.0292	-.0137	-.0018	-.0225	-.0766
	133	-.0851	-.0653	-.0517	-.0355	-.0247	-.0056	-.0392	-.0952
	134	-.1268	-.0918	-.0727	-.0549	-.0434	-.0176	-.0319	-.0916
	135	-.1535	-.1186	-.0942	-.0698	-.0451	-.0193	-.0305	-.0739
	136	-.1834	-.1530	-.1200	-.0843	-.0546	-.0316	-.0347	-.0620
	137	-.0909	-.0826	-.0752	-.0838	-.0780	-.0666	-.0854	-.0912
	138	-.1488	-.1323	-.1151	-.1085	-.0950	-.0831	-.1038	-.1093
	139	-.1570	-.1430	-.1274	-.1117	-.0980	-.0765	-.1022	-.1054
	140	-.1760	-.1535	-.1361	-.1213	-.1007	-.0872	-.1046	-.0980
	141	-.1921	-.1711	-.1560	-.1331	-.1114	-.0863	-.0956	-.0931
	142	-.2069	-.1832	-.1617	-.1386	-.1106	-.0844	-.0867	-.0876
	143	.0217	.0142	-.0028	-.0187	-.0385	-.0400	-.0450	-.0541
	144	-.0156	-.0204	-.0311	-.0447	-.0436	-.0368	-.0458	-.0525
	145	-.0237	-.0311	-.0354	-.0296	-.0327	-.0242	-.0439	-.0437
	146	-.0203	-.0223	-.0176	-.0247	-.0230	-.0217	-.0418	-.0245
	147	-.0716	-.0703	-.0733	-.0607	-.0603	-.0530	-.0407	-.0346

TABLE X - Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(d) $M = 0.95$; $\beta = -5.8^{\circ}$.

	Orifice number	C_p at α, deg, of —								
		-8.06	-7.45	-4.95	-2.39	0.31	2.86	5.41	7.93	9.26
Fore fuselage area-rule addition	101	.3687	.3626	.3291	.2913	.2464	.2158	.1983	.1759	.1614
	102	.2830	.2676	.2303	.1834	.1384	.1071	.0720	.0528	.0415
	103	-.0216	-.0226	-.0044	.0084	.0094	.0230	.0356	.0373	.0349
	104	-.0436	-.0289	-.0066	.0140	.0424	.0759	.1056	.1393	.1574
	105	.0005	-.0037	-.0098	-.0240	-.0466	-.0521	-.0576	-.0656	-.0700
	106	-.1463	-.1387	-.0912	-.0652	-.0579	-.0419	-.0301	-.0284	-.0209
	107	-.2793	-.2711	-.1360	-.0833	-.0286	.0152	.0562	.0951	.1094
	108	-.0306	-.0438	-.0786	-.1208	-.1603	-.1806	-.2704	-.3037	-.3117
	109	-.1625	-.1586	-.1536	-.1462	-.1564	-.1688	-.1901	-.1916	-.1881
	110	-.1959	-.1972	-.1732	-.1605	-.1525	-.1519	-.1598	-.1515	-.1520
	111	-.2077	-.2169	-.1671	-.1345	-.1134	-.1001	-.0874	-.0724	-.0690
	112	-.2571	-.2377	-.1998	-.1436	-.1041	-.0720	-.0330	-.0005	.0221
	113	-.0458	-.0431	-.0375	-.0322	-.0194	-.0054	.0126	.0407	.0551
	114	-.0954	-.0974	-.0883	-.0829	-.0770	-.0715	-.0688	-.0697	-.0676
	115	-.1992	-.1951	-.1512	-.1370	-.1279	-.1098	-.1034	-.1053	-.1061
	116	-.2445	-.2529	-.1979	-.1840	-.1622	-.1407	-.1254	-.1147	-.1077
	117	-.1571	-.1588	-.1296	-.0895	-.0468	.0065	.0626	.1220	.1512
	118	-.2503	-.2381	-.1749	-.1545	-.1129	-.0698	-.0356	.0000	.0155
	119	-.2992	-.2938	-.2102	-.1889	-.1462	-.1059	-.0773	-.0519	-.0402
	120	-.2612	-.2352	-.1846	-.1661	-.1249	-.0977	-.0731	-.0540	-.0468
	121	-.1944	-.1806	-.1894	-.1601	-.1184	-.0889	-.0625	-.0429	-.0270
	122	-.1088	-.1060	-.0798	-.0475	-.0057	.0481	.0919	.1672	.1958
	123	-.0998	-.0954	-.0620	-.0272	.0211	.0633	.1165	.1768	.2045
	124	-.0846	-.0775	-.0540	-.0180	.0189	.0604	.1162	.1596	.1917
	125	-.0589	-.0524	-.0245	-.0069	.0301	.0674	.1024	.1455	.1737
	126	-.0562	-.0477	-.0269	.0068	.0337	.0680	.0958	.1337	.1541
Aft fuselage area-rule addition	127	.0865	.0900	.1141	.1635	.1770	.1819	.1900	.1983	.2033
	128	.0370	.0417	.0763	.1185	.1443	.1630	.1764	.1993	.2093
	129	.0759	.0788	.0799	.0929	.0845	.0719	.0425	.0317	.0240
	130	-.1765	-.1734	-.1217	-.0773	-.0383	-.0173	.0028	.0210	.0324
	131	-.0439	-.0378	-.0146	-.0114	-.0199	-.0059	-.0151	-.0344	-.0584
	132	-.0976	-.0870	-.0584	-.0354	-.0313	-.0199	-.0291	-.0371	-.0656
	133	-.1185	-.1060	-.0780	-.0494	-.0364	-.0233	-.0296	-.0528	-.0683
	134	-.1501	-.1377	-.1038	-.0773	-.0577	-.0459	-.0477	-.0480	-.0707
	135	-.1720	-.1567	-.1237	-.0918	-.0693	-.0425	-.0397	-.0421	-.0526
	136	-.2145	-.2046	-.1599	-.1191	-.0904	-.0584	-.0500	-.0438	-.0485
	137	-.1092	-.1066	-.0910	-.0855	-.0931	-.0761	-.0669	-.0727	-.0790
	138	-.1739	-.1700	-.1493	-.1322	-.1231	-.0976	-.0824	-.0991	-.1065
	139	-.1814	-.1777	-.1578	-.1392	-.1223	-.0982	-.0846	-.1005	-.1039
	140	-.2041	-.2028	-.1754	-.1516	-.1352	-.1011	-.0928	-.0923	-.1017
	141	-.2238	-.2217	-.1935	-.1673	-.1524	-.1126	-.0925	-.1019	-.0998
	142	-.2339	-.2328	-.2052	-.1848	-.1570	-.1206	-.0937	-.0925	-.0945
	143	.0229	.0295	.0174	.0105	-.0153	-.0364	-.0349	-.0354	-.0470
	144	.0004	-.0015	-.0022	-.0187	-.0434	-.0398	-.0301	-.0320	-.0400
	145	.0051	-.0011	-.0115	-.0177	-.0238	-.0241	-.0216	-.0395	-.0453
	146	.0069	.0119	.0035	.0030	-.0156	-.0134	-.0180	-.0243	-.0177
	147	-.0403	-.0393	-.0388	-.0400	-.0460	-.0446	-.0489	-.0344	-.0388

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(e) M = 0.98; β = -5.8°.

	Orifice number	C_p at α, deg, of —						
		-5.00	-2.24	0.27	2.93	5.57	7.85	9.79
Fore fuselage area-rule addition	101	.3462	.3106	.2653	.2424	.2193	.1932	.1676
	102	.2426	.1949	.1499	.1269	.0956	.0731	.0532
	103	.0195	.0231	.0241	.0456	.0551	.0576	.0486
	104	.0192	.0459	.0595	.0951	.1271	.1553	.1753
	105	-.0181	-.0232	-.0287	-.0311	-.0357	-.0437	-.0590
	106	-.1236	-.0698	-.0427	-.0225	-.0116	-.0101	-.0080
	107	-.2142	-.0646	-.0213	.0333	.0787	.1041	.1282
	108	-.0859	-.1339	-.1874	-.2293	-.2566	-.2786	-.2996
	109	-.1149	-.1404	-.1738	-.1704	-.1704	-.1755	-.1819
	110	-.1127	-.1566	-.1602	-.1455	-.1415	-.1368	-.1386
	111	-.1142	-.1301	-.1261	-.0921	-.0735	-.0645	-.0614
	112	-.1536	-.1369	-.1107	-.0614	-.0259	.0063	.0350
	113	-.0147	-.0133	-.0051	.0182	.0423	.0735	.0979
	114	-.0680	-.0645	-.0566	-.0440	-.0652	-.0594	-.0312
	115	-.1435	-.1293	-.1099	-.1917	-.2728	-.3034	-.3199
	116	-.1965	-.1652	-.1431	-.2435	-.2481	-.2415	-.2333
	117	-.1299	-.0877	-.0543	.0143	.0821	.1344	.1807
	118	-.2013	-.1523	-.1174	-.0675	-.0120	.0253	.0459
	119	-.2333	-.1859	-.1561	-.1097	-.0525	-.0294	-.0166
	120	-.2086	-.1656	-.1416	-.1018	-.0628	-.0374	-.0295
	121	-.1890	-.1518	-.1325	-.0974	-.0554	-.0295	-.0188
	122	-.0573	-.0263	.0059	.0485	.0981	.1567	.2149
	123	-.0392	-.0074	.0306	.0815	.1364	.1815	.2277
	124	-.0394	.0044	.0335	.0769	.1296	.1771	.2204
	125	-.0126	.0166	.0388	.0811	.1245	.1590	.1945
	126	-.0087	.0241	.0531	.0849	.1180	.1563	.1760
Aft fuselage area-rule addition	127	.1265	.1603	.1928	.2034	.2125	.2149	.2242
	128	.0844	.1284	.1589	.1692	.1910	.2167	.2278
	129	.0837	.0966	.1121	.0791	.0559	.0460	.0277
	130	-.1452	-.0711	-.0173	-.0089	.0110	.0325	.0478
	131	-.0343	-.0133	-.0009	-.0195	-.0346	-.0459	-.0842
	132	-.0786	-.0348	-.0157	-.0302	-.0363	-.0481	-.0808
	133	-.0972	-.0534	-.0277	-.0351	-.0406	-.0558	-.0874
	134	-.1215	-.0720	-.0413	-.0517	-.0576	-.0536	-.0885
	135	-.1377	-.0935	-.0566	-.0576	-.0495	-.0478	-.0646
	136	-.1793	-.1202	-.0708	-.0631	-.0541	-.0455	-.0503
	137	-.1193	-.1006	-.0991	-.0900	-.1216	-.1402	-.1505
	138	-.1870	-.1391	-.1225	-.1142	-.1386	-.1644	-.1764
	139	-.1890	-.1501	-.1257	-.1136	-.1359	-.1565	-.1641
	140	-.2129	-.1618	-.1385	-.1199	-.1393	-.1547	-.1522
	141	-.2408	-.1819	-.1540	-.1366	-.1483	-.1541	-.1539
	142	-.2522	-.1976	-.1637	-.1442	-.1471	-.1506	-.1425
	143	.0440	.0282	.0024	-.0190	-.0275	-.0362	-.0702
	144	.0201	.0007	-.0254	-.0236	-.0196	-.0444	-.0755
	145	.0242	.0039	-.0005	-.0099	-.0103	-.0527	-.0827
	146	.0313	.0247	.0038	-.0021	-.0157	-.0445	-.0668
	147	-.0063	-.0106	-.0265	-.0343	-.0466	-.0513	-.1062

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(f) M = 0.99; $\beta = -5.8^{\circ}$.

	Orifice number	C_p at α, deg, of —						
		-4.88	-2.35	0.31	2.95	5.52	7.86	9.80
Fore fuselage area-rule addition	101	.3561	.3189	.2784	.2461	.2269	.2013	.1817
	102	.2565	.2076	.1632	.1345	.1060	.0822	.0616
	103	.0388	.0337	.0378	.0584	.0649	.0674	.0563
	104	.0317	.0541	.0658	.1032	.1333	.1644	.1868
	105	-.0031	-.0249	-.0171	-.0203	-.0234	-.0335	-.0464
	106	-.1120	-.0816	-.0304	-.0103	-.0019	-.0006	.0033
	107	-.1978	-.1213	-.0071	.0474	.0822	.1168	.1421
	108	-.1500	-.1610	-.1757	-.2145	-.2452	-.2678	-.2860
	109	-.2120	-.1679	-.1624	-.1594	-.1613	-.1654	-.1665
	110	-.2325	-.1719	-.1491	-.1367	-.1247	-.1246	-.1261
	111	-.1614	-.1416	-.1094	-.0812	-.0622	-.0546	-.0466
	112	-.1874	-.1456	-.0900	-.0494	-.0126	.0213	.0461
	113	-.0175	-.0065	.0030	.0018	.0136	.0472	.0859
	114	-.0626	-.0477	-.0532	-.1255	-.2317	-.3102	-.2394
	115	-.1370	-.1162	-.1586	-.2401	-.2707	-.2889	-.3095
	116	-.1929	-.1565	-.2164	-.2395	-.2379	-.2334	-.2244
	117	-.1304	-.0872	-.0466	.0149	.0853	.1411	.1846
	118	-.1847	-.1439	-.1082	-.0661	-.0191	.0417	.0665
	119	-.2212	-.1819	-.1341	-.1024	-.0600	-.0135	.0028
	120	-.1951	-.1593	-.1239	-.0974	-.0639	-.0130	-.0088
	121	-.1726	-.1458	-.1101	-.0928	-.0698	-.0186	-.0084
	122	-.0469	-.0345	.0089	.0475	.0873	.1282	.2089
	123	-.0364	-.0130	.0366	.0841	.1388	.1877	.2276
	124	-.0376	-.0065	.0376	.0827	.1366	.1802	.2320
	125	-.0102	.0076	.0494	.0876	.1328	.1720	.2057
	126	-.0120	.0208	.0529	.0944	.1264	.1663	.1985
Aft fuselage area-rule addition	127	.1383	.1598	.1837	.2045	.2242	.2297	.2388
	128	.1028	.1277	.1432	.1698	.1916	.2196	.2388
	129	.0959	.1011	.1109	.0808	.0616	.0569	.0480
	130	-.1329	-.0680	-.0163	-.0065	.0156	.0411	.0602
	131	-.0255	-.0067	.0165	-.0080	-.0227	-.0345	-.0670
	132	-.0690	-.0310	-.0009	-.0223	-.0295	-.0379	-.0639
	133	-.0875	-.0465	-.0115	-.0271	-.0278	-.0435	-.0696
	134	-.1103	-.0673	-.0307	-.0443	-.0508	-.0571	-.0694
	135	-.1241	-.0840	-.0442	-.0538	-.0451	-.0384	-.0508
	136	-.1678	-.1080	-.0585	-.0601	-.0438	-.0404	-.0386
	137	-.1147	-.0949	-.0823	-.0984	-.1132	-.1293	-.1355
	138	-.1774	-.1335	-.1029	-.1183	-.1294	-.1514	-.1636
	139	-.1704	-.1438	-.1079	-.1165	-.1246	-.1422	-.1542
	140	-.2123	-.1545	-.1181	-.1238	-.1286	-.1520	-.1404
	141	-.2366	-.1800	-.1314	-.1371	-.1439	-.1431	-.1394
	142	-.2523	-.1864	-.1423	-.1379	-.1405	-.1385	-.1327
	143	.0131	.0463	.0092	-.0133	-.1143	-.1898	-.2386
	144	.0447	.0023	-.0164	-.0182	-.1047	-.1758	-.2228
	145	.0459	.0147	.0040	-.0082	-.0967	-.1758	-.2078
	146	.0337	.0350	.0095	-.0031	-.1115	-.1509	-.1785
	147	.0159	-.0031	-.0255	-.0429	-.1447	-.1525	-.1906

TABLE X - Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(g) $M = 1.00$; $\beta = -5.7^{o}$.

	Orifice number	C_p at α, deg, of —						
		-2.97	-2.28	0.28	2.85	5.38	7.84	9.78
Fore fuselage area-rule addition	101	.3395	.3324	.2817	.2591	.2304	.2042	.1835
	102	.2245	.2091	.1697	.1417	.1135	.0891	.0694
	103	.0532	.0456	.0493	.0626	.0684	.0675	.0656
	104	.0668	.0704	.0794	.1114	.1370	.1643	.1878
	105	-.0089	-.0135	-.0150	-.0101	-.0206	-.0305	-.0375
	106	-.0981	-.0668	-.0263	-.0014	.0051	.0020	.0080
	107	-.1303	-.1059	.0056	.0509	.0903	.1182	.1374
	108	-.1430	-.1476	-.1691	-.2054	-.2326	-.2587	-.2729
	109	-.1476	-.1574	-.1475	-.1511	-.1501	-.1526	-.1621
	110	-.1560	-.1552	-.1377	-.1275	-.1214	-.1187	-.1197
	111	-.1343	-.1328	-.0996	-.0742	-.0580	-.0498	-.0433
	112	-.1519	-.1396	-.0832	-.0389	-.0072	.0260	.0466
	113	-.0147	-.0111	-.0014	-.0024	-.0002	.0134	.0473
	114	-.0808	-.0639	-.0687	-.1664	-.2686	-.3226	-.3466
	115	-.1644	-.1613	-.2232	-.2360	-.2553	-.2812	-.2990
	116	-.2087	-.2073	-.2305	-.2308	-.2341	-.2263	-.2138
	117	-.0806	-.0731	-.0310	.0131	.0583	.1271	.1840
	118	-.1557	-.1396	-.1069	-.0680	-.0450	.0275	.0764
	119	-.1850	-.1690	-.1325	-.1110	-.0844	-.0305	.0211
	120	-.1724	-.1396	-.1188	-.0979	-.0861	-.0432	.0056
	121	-.1495	-.1391	-.1086	-.0907	-.0873	-.0783	-.0044
	122	-.0315	-.0194	-.0089	.0245	.0760	.1135	.2017
	123	-.0165	-.0106	.0194	.0658	.1271	.1866	.2271
	124	-.0104	-.0021	.0281	.0660	.1300	.1892	.2403
	125	.0088	.0158	.0391	.0742	.1293	.1791	.2167
	126	.0141	.0247	.0379	.0789	.1257	.1667	.2027
Aft fuselage area-rule addition	127	.1661	.1680	.1754	.2189	.2282	.2344	.2418
	128	.1243	.1293	.1325	.1766	.1913	.2203	.2395
	129	.1114	.1021	.0906	.0815	.0628	.0516	.0538
	130	-.0908	-.0665	-.0373	.0015	.0184	.0449	.0641
	131	-.0044	-.0084	.0025	-.0159	-.0252	-.0365	-.0648
	132	-.0321	-.0312	-.0208	-.0275	-.0312	-.0355	-.0551
	133	-.0488	-.0442	-.0317	-.0302	-.0298	-.0401	-.0634
	134	-.0680	-.0653	-.0496	-.0443	-.0457	-.0564	-.0677
	135	-.0803	-.0839	-.0581	-.0459	-.0414	-.0380	-.0438
	136	-.1152	-.1088	-.0805	-.0612	-.0454	-.0348	-.0361
	137	-.0880	-.0822	-.0649	-.0871	-.1057	-.1204	-.1321
	138	-.1354	-.1250	-.0934	-.1081	-.1217	-.1429	-.1572
	139	-.1398	-.1299	-.0990	-.1047	-.1151	-.1309	-.1474
	140	-.1551	-.1422	-.1031	-.1105	-.1217	-.1449	-.1398
	141	-.1776	-.1594	-.1217	-.1258	-.1364	-.1347	-.1393
	142	-.1873	-.1708	-.1299	-.1338	-.1361	-.1333	-.1273
	143	-.1636	-.0510	-.0010	-.1868	-.2387	-.2600	-.2731
	144	-.0409	-.0140	-.0211	-.1420	-.2138	-.2411	-.2543
	145	-.0385	-.0191	.0002	-.0587	-.1841	-.2261	-.2419
	146	-.0114	-.0002	.0105	-.0898	-.1739	-.2032	-.1945
	147	-.0673	-.0390	-.0261	-.1260	-.1853	-.1809	-.2035

[redacted]

TABLE X - Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(h) M = 0.50; $\beta = 0^{o}$.

	Orifice number	C_p at α, deg, of —														
		-6.58	-5.43	-3.23	-1.00	0.09	1.19	2.30	3.39	4.50	5.61	7.79	10.01	12.14	1323	1431
Fore fuselage area-rule addition	101	.1264	.1205	.1040	.0794	.0624	.0483	.0408	.0360	.0109	.0122	-.0267	-.0724	-.1700	-.1997	-.2548
	102	.1047	.0957	.0698	.0360	.0228	.0112	-.0046	-.0197	-.0301	-.0456	-.0718	-.0996	-.1232	-.1280	-.1423
	103	-.1752	-.1636	-.1407	-.1256	-.1239	-.1162	-.1099	-.1092	-.1037	-.1028	-.0989	-.0752	-.0694	-.0772	-.0924
	104	-.1610	-.1510	-.1061	-.0919	-.0672	-.0582	-.0475	-.0392	-.0204	-.0096	.0203	.0381	.0422	.0342	.0115
	105	-.0714	-.0738	-.0744	-.0852	-.0893	-.0941	-.0998	-.1035	-.1135	-.1154	-.1346	-.1506	-.1694	-.1817	-.2017
	106	-.1759	-.1592	-.1307	-.1189	-.1135	-.1118	-.1036	-.0953	-.0949	-.0981	-.1071	-.1177	-.1478	-.1599	-.1796
	107	-.1838	-.1636	-.1184	-.0796	-.0644	-.0519	-.0443	-.0285	-.0191	-.0153	-.0077	-.0049	-.0179	-.0257	-.0321
	108	-.0979	-.1028	-.1090	-.1218	-.1268	-.1329	-.1395	-.1426	-.1533	-.1562	-.1700	-.1848	-.2041	-.2101	-.2254
	109	-.1980	-.1948	-.1719	-.1558	-.1570	-.1493	-.1493	-.1460	-.1473	-.1489	-.1640	-.1791	-.2057	-.2158	-.2356
	110	-.2185	-.2045	-.1747	-.1631	-.1561	-.1493	-.1471	-.1441	-.1438	-.1480	-.1583	-.1825	-.2089	-.2228	-.2495
	111	-.2033	-.1897	-.1527	-.1322	-.1255	-.1197	-.1168	-.1082	-.1085	-.1082	-.1201	-.1363	-.1659	-.1704	-.1913
	112	-.1894	-.1708	-.1392	-.1123	-.1003	-.0960	-.0831	-.0764	-.0728	-.0665	-.0611	-.0572	-.0688	-.0734	-.0785
	113	-.1080	-.1034	-.0926	-.0805	-.0780	-.0777	-.0724	-.0638	-.0636	-.0595	-.0573	-.0553	-.0656	-.0661	-.0753
	114	-.1790	-.1652	-.1473	-.1379	-.1271	-.1272	-.1212	-.1170	-.1173	-.1176	-.1258	-.1363	-.1589	-.1675	-.1724
	115	-.2222	-.2092	-.1737	-.1527	-.1466	-.1420	-.1392	-.1347	-.1328	-.1344	-.1460	-.1696	-.2010	-.2136	-.2314
	116	-.2238	-.2020	-.1800	-.1605	-.1482	-.1496	-.1462	-.1413	-.1438	-.1480	-.1580	-.1759	-.2013	-.2105	-.2311
	117	-.1980	-.1715	-.1432	-.1076	-.0975	-.0739	-.0585	-.0430	-.0232	-.0049	.0115	.0223	.0422	.0619	.0673
	118	-.2200	-.1951	-.1609	-.1297	-.1183	-.0960	-.0982	-.0855	-.0772	-.0766	-.0702	-.0793	-.0878	-.0785	-.0956
	119	-.2191	-.2007	-.1643	-.1404	-.1198	-.1152	-.1134	-.1063	-.0999	-.0984	-.1084	-.1120	-.1342	-.1410	-.1594
	120	-.1705	-.1598	-.1307	-.1136	-.1050	-.1045	-.1001	-.0985	-.0974	-.0996	-.1116	-.1332	-.1532	-.1580	-.1828
	121	-.1434	-.1274	-.1109	-.1000	-.0968	-.0947	-.0913	-.0903	-.0930	-.0984	-.1068	-.1237	-.1513	-.1634	-.1724
	122	-.1797	-.1289	-.1134	-.0670	-.0364	-.0225	-.0071	.0134	.0342	.0550	.1009	.1213	.1258	.1366	.1285
	123	-.1582	-.1277	-.0920	-.0518	-.0320	-.0099	.0096	.0253	.0402	.0601	.0811	.1213	.1315	.1423	.1477
	124	-.1371	-.1056	-.0719	-.0395	-.0222	-.0093	.0046	.0253	.0248	.0456	.0669	.0852	.1144	.1237	.1294
	125	-.0945	-.0748	-.0492	-.0269	-.0083	-.0017	.0096	.0197	.0270	.0358	.0439	.0593	.0668	.0755	.0749
	126	-.0759	-.0631	-.0341	-.0172	-.0043	.0033	.0093	.0159	.0185	.0263	.0279	.0233	.0191	.0207	.0159
Aft fuselage area-rule addition	127	.0777	.0817	.0876	.1018	.1084	.1105	.1270	.1301	.1355	.1451	.1580	.1449	.1287	.1155	.0986
	128	.0430	.0449	.0635	.0808	.0874	.0968	.1022	.1078	.1173	.1240	.1395	.1557	.1481	.1343	.1394
	129	.0386	.0334	.0337	.0335	.0334	.0341	.0408	.0443	.0460	.0485	.0631	.0647	.0594	.0448	.0502
	130	-.0885	-.0836	-.0561	-.0383	-.0263	-.0172	-.0030	.0024	.0090	.0170	.0383	.0448	.0521	.0358	.0498
	131	-.0241	-.0269	-.0272	-.0320	-.0301	-.0279	-.0207	-.0187	-.0195	-.0106	-.0068	-.0112	-.0156	-.0314	-.0317
	132	-.0500	-.0505	-.0442	-.0405	-.0361	-.0367	-.0295	-.0276	-.0279	-.0279	-.0229	-.0106	-.0160	-.0232	-.0283
	133	-.0626	-.0603	-.0486	-.0430	-.0424	-.0396	-.0285	-.0260	-.0219	-.0223	-.0140	-.0283	-.0545	-.0661	-.0782
	134	-.0856	-.0855	-.0684	-.0591	-.0483	-.0443	-.0411	-.0307	-.0301	-.0286	-.0125	-.0248	-.0485	-.0661	-.0690
	135	-.0951	-.0937	-.0706	-.0603	-.0556	-.0538	-.0370	-.0339	-.0283	-.0241	-.0099	-.0096	-.0245	-.0368	-.0314
	136	-.1049	-.1019	-.0816	-.0644	-.0575	-.0493	-.0430	-.0345	-.0283	-.0226	-.0071	.0002	-.0020	-.0096	-.0027
	137	-.0819	-.0861	-.0797	-.0783	-.0779	-.0793	-.0695	-.0688	-.0712	-.0737	-.0633	-.0644	-.0760	-.0838	-.0861
	138	-.1115	-.1091	-.1027	-.0966	-.0934	-.0951	-.0884	-.0887	-.0873	-.0964	-.0828	-.0815	-.0881	-.1053	-.1038
	139	-.1143	-.1138	-.0983	-.0915	-.0874	-.0884	-.0790	-.0739	-.0728	-.0731	-.0759	-.0822	-.0991	-.1258	-.1227
	140	-.1156	-.1141	-.0989	-.0871	-.0773	-.0783	-.0720	-.0660	-.0633	-.0633	-.0551	-.0701	-.0745	-.0683	-.0655
	141	-.1263	-.1160	-.0980	-.0922	-.0836	-.0831	-.0661	-.0654	-.0639	-.0557	-.0462	-.0575	-.0599	-.0680	-.0583
	142	-.1203	-.1126	-.0958	-.0824	-.0773	-.0758	-.0616	-.0606	-.0589	-.0516	-.0446	-.0356	-.0349	-.0497	-.0475
	143	-.0396	-.0424	-.0448	-.0543	-.0546	-.0619	-.0550	-.0619	-.0674	-.0677	-.0787	-.0917	-.1020	-.1148	-.1107
	144	-.0509	-.0509	-.0445	-.0496	-.0455	-.0509	-.0531	-.0540	-.0567	-.0530	-.0677	-.0765	-.0912	-.1091	-.1120
	145	-.0494	-.0458	-.0451	-.0430	-.0398	-.0392	-.0389	-.0386	-.0393	-.0453	-.0519	-.0679	-.0741	-.0848	-.0687
	146	-.0626	-.0616	-.0486	-.0358	-.0329	-.0304	-.0254	-.0273	-.0229	-.0260	-.0254	-.0274	-.0232	-.0421	-.0400
	147	-.1099	-.1059	-.0794	-.0635	-.0565	-.0522	-.0411	-.0392	-.0371	-.0320	-.0197	-.0248	-.0232	-.0396	-.0305

[redacted]

[redacted]

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(i) $M = 0.80$; $\beta = 0^{\circ}$.

	Orifice number	C_p at α, deg, of —														
		-7.42	-6.18	-4.95	-3.63	-2.30	-1.01	0.25	1.54	2.83	4.14	5.42	6.71	7.93	9.07	10.24
Fore fuselage area-rule addition	101	.1595	.1477	.1258	.1201	.1050	.0903	.0705	.0570	.0470	.0278	.0225	-.0031	-.0422	-.0628	-.1299
	102	.1200	.1095	.0957	.0729	.0613	.0336	.0217	.0092	-.0039	-.0246	-.0411	-.0591	-.0724	-.0729	-.0769
	103	-.2223	-.2016	-.1894	-.1644	-.1562	-.1537	-.1471	-.1322	-.1274	-.1098	-.0990	-.0963	-.0814	-.0522	-.0311
	104	-.2042	-.1905	-.1544	-.1314	-.1007	-.0948	-.0724	-.0608	-.0488	-.0320	-.0075	.0261	.0614	.0900	.1052
	105	-.0899	-.0891	-.0906	-.0969	-.1004	-.1069	-.1122	-.1188	-.1247	-.1290	-.1355	-.1424	-.1509	-.1497	-.1602
	106	-.2053	-.1814	-.1674	-.1531	-.1463	-.1375	-.1317	-.1215	-.1182	-.1123	-.1122	-.1140	-.1146	-.1116	-.1266
	107	-.2289	-.1981	-.1638	-.1367	-.1119	-.0902	-.0699	-.0570	-.0375	-.0234	-.0133	-.0089	-.0004	.0048	.0009
	108	-.1091	-.1114	-.1195	-.1301	-.1411	-.1457	-.1522	-.1605	-.1653	-.1740	-.1842	-.1929	-.1968	-.1991	-.2080
	109	-.2379	-.2242	-.2099	-.1940	-.1859	-.1830	-.1806	-.1735	-.1744	-.1702	-.1719	-.1770	-.1827	-.1890	-.2003
	110	-.2520	-.2368	-.2089	-.1998	-.1881	-.1831	-.1766	-.1724	-.1645	-.1672	-.1705	-.1759	-.1755	-.1827	-.2068
	111	-.2363	-.2176	-.1918	-.1759	-.1606	-.1485	-.1435	-.1309	-.1278	-.1251	-.1256	-.1243	-.1311	-.1358	-.1462
	112	-.2141	-.1957	-.1773	-.1647	-.1455	-.1294	-.1187	-.1057	-.0937	-.0856	-.0778	-.0732	-.0707	-.0603	-.0605
	113	-.1209	-.1133	-.1084	-.1023	-.0938	-.0913	-.0863	-.0803	-.0721	-.0639	-.0586	-.0501	-.0474	-.0407	-.0419
	114	-.1932	-.1800	-.1712	-.1634	-.1501	-.1413	-.1396	-.1298	-.1283	-.1267	-.1221	-.1243	-.1280	-.1311	-.1368
	115	-.2487	-.2272	-.2023	-.1878	-.1781	-.1688	-.1635	-.1521	-.1571	-.1504	-.1579	-.1572	-.1709	-.1689	-.1849
	116	-.2508	-.2325	-.2103	-.1966	-.1910	-.1835	-.1722	-.1697	-.1698	-.1620	-.1564	-.1704	-.1803	-.1871	-.2008
	117	-.2314	-.2156	-.1879	-.1639	-.1487	-.1284	-.1107	-.0800	-.0621	-.0424	-.0210	.0030	.0214	.0285	.0362
	118	-.2560	-.2262	-.2074	-.1848	-.1631	-.1474	-.1291	-.1164	-.1055	-.0899	-.0829	-.0770	-.0759	-.0690	-.0720
	119	-.2550	-.2299	-.2036	-.1949	-.1756	-.1588	-.1456	-.1367	-.1256	-.1193	-.1138	-.1088	-.1127	-.1113	-.1162
	120	-.2041	-.1827	-.1693	-.1570	-.1427	-.1312	-.1278	-.1194	-.1141	-.1122	-.1122	-.1136	-.1213	-.1239	-.1355
	121	-.1619	-.1460	-.1335	-.1274	-.1243	-.1158	-.1129	-.1089	-.1064	-.1072	-.1095	-.1168	-.1245	-.1242	-.1358
	122	-.1942	-.1618	-.1340	-.1224	-.0897	-.0690	-.0449	-.0155	.0090	.0239	.0533	.0823	.1077	.1297	.1417
	123	-.1564	-.1346	-.1091	-.0924	-.0653	-.0535	-.0212	-.0064	.0173	.0362	.0566	.0784	.0927	.1093	.1282
	124	-.1374	-.1083	-.0908	-.0762	-.0551	-.0408	-.0152	-.0065	.0156	.0256	.0413	.0597	.0778	.1063	.1183
	125	-.0951	-.0784	-.0688	-.0473	-.0325	-.0230	-.0020	.0026	.0136	.0231	.0346	.0475	.0534	.0789	.0850
	126	-.0768	-.0644	-.0475	-.0353	-.0221	-.0125	-.0072	.0026	.0173	.0191	.0294	.0380	.0404	.0470	.0475
Aft fuselage area-rule addition	127	.0955	.1016	.1008	.1049	.1198	.1229	.1348	.1432	.1499	.1581	.1735	.1813	.1775	.1737	.1659
	128	.0518	.0594	.0721	.0827	.0902	.0947	.1110	.1194	.1306	.1441	.1486	.1553	.1605	.1546	.1567
	129	.0345	.0425	.0410	.0437	.0432	.0392	.0453	.0480	.0471	.0550	.0621	.0621	.0620	.0466	.0353
	130	-.0968	-.0916	-.0736	-.0631	-.0509	-.0391	-.0235	-.0089	-.0040	.0141	.0244	.0271	.0415	.0348	.0359
	131	-.0435	-.0348	-.0345	-.0317	-.0312	-.0356	-.0333	-.0287	-.0292	-.0232	-.0169	-.0188	-.0256	-.0350	-.0433
	132	-.0701	-.0605	-.0524	-.0477	-.0443	-.0495	-.0411	-.0353	-.0355	-.0309	-.0262	-.0246	-.0328	-.0448	-.0345
	133	-.0841	-.0743	-.0681	-.0571	-.0546	-.0507	-.0425	-.0405	-.0347	-.0292	-.0229	-.0226	-.0347	-.0440	-.0553
	134	-.1054	-.0910	-.0794	-.0735	-.0697	-.0658	-.0548	-.0490	-.0437	-.0331	-.0268	-.0234	-.0304	-.0573	-.0715
	135	-.1111	-.1009	-.0942	-.0942	-.0740	-.0719	-.0624	-.0427	-.0418	-.0356	-.0265	-.0224	-.0259	-.0322	-.0388
	136	-.1232	-.1135	-.0950	-.0957	-.0815	-.0735	-.0636	-.0556	-.0418	-.0334	-.0224	-.0234	-.0168	-.0172	-.0314
	137	-.1144	-.1076	-.1017	-.0993	-.0944	-.0966	-.0907	-.0854	-.0849	-.0815	-.0834	-.0822	-.0822	-.0853	-.0806
	138	-.1372	-.1347	-.1242	-.1252	-.1229	-.1199	-.1133	-.1101	-.1083	-.1023	-.0994	-.1048	-.1036	-.1014	-.1001
	139	-.1443	-.1377	-.1313	-.1213	-.1117	-.1144	-.1067	-.0993	-.0978	-.0935	-.0869	-.0952	-.0968	-.0932	-.1014
	140	-.1482	-.1369	-.1313	-.1197	-.1097	-.1076	-.0999	-.0917	-.0857	-.0796	-.0727	-.0743	-.0798	-.0943	-.1015
	141	-.1443	-.1366	-.1247	-.1169	-.1128	-.1057	-.0977	-.0875	-.0875	-.0799	-.0752	-.0737	-.0781	-.0715	-.0808
	142	-.1386	-.1339	-.1244	-.1150	-.1086	-.1054	-.0924	-.0864	-.0847	-.0752	-.0664	-.0661	-.0657	-.0594	-.0660
	143	-.0603	-.0580	-.0501	-.0625	-.0633	-.0737	-.0715	-.0738	-.0784	-.0809	-.0853	-.0885	-.0954	-.0930	-.0956
	144	-.0686	-.0622	-.0601	-.0625	-.0645	-.0710	-.0672	-.0671	-.0695	-.0737	-.0760	-.0776	-.0863	-.0778	-.0784
	145	-.0518	-.0446	-.0417	-.0518	-.0564	-.0564	-.0526	-.0586	-.0553	-.0564	-.0559	-.0628	-.0635	-.0680	-.0778
	146	-.0476	-.0495	-.0549	-.0529	-.0498	-.0510	-.0430	-.0361	-.0336	-.0350	-.0381	-.0366	-.0407	-.0356	-.0306
	147	-.0949	-.0874	-.0813	-.0773	-.0693	-.0664	-.0592	-.0498	-.0514	-.0461	-.0427	-.0427	-.0386	-.0383	-.0339

[redacted]

[redacted]

TABLE X - Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(j) $M = 0.90;\ \beta = 0^{o}$.

	Orifice number	C_p at α, deg, of —											
		-6.35	-5.05	-3.76	-2.39	-1.06	0.25	1.60	2.88	4.25	5.60	6.87	8.01
Fore fuselage area-rule addition	101	.1648	.1476	.1040	.0794	.0991	.0877	.0644	.0569	.0423	.0262	.0030	-.0500
	102	.1172	.1004	.0894	.0696	.0482	.0363	.0198	.0074	-.0105	-.0252	-.0351	-.0505
	103	-.2111	-.1992	-.1810	-.1687	-.1530	-.1400	-.1333	-.1186	-.1080	-.1012	-.0760	-.0587
	104	-.2011	-.1650	-.1327	-.1231	-.0977	-.0724	-.0533	-.0371	-.0116	.0189	.0555	.1115
	105	-.1070	-.1129	-.1135	-.1163	-.1218	-.1269	-.1331	-.1375	-.1407	-.1487	-.1496	-.1555
	106	-.1900	-.1775	-.1675	-.1531	-.1472	-.1385	-.1328	-.1233	-.1217	-.1182	-.1164	-.1135
	107	-.1963	-.1683	-.1475	-.1267	-.0985	-.0785	-.0604	-.0392	-.0242	-.0126	.0002	.0041
	108	-.1201	-.1362	-.1453	-.1567	-.1628	-.1697	-.1782	-.1863	-.1947	-.2038	-.2121	-.2191
	109	-.2342	-.2294	-.2166	-.2030	-.1974	-.1951	-.1934	-.1913	-.1939	-.1907	-.1948	-.2042
	110	-.2394	-.2271	-.2130	-.2047	-.1928	-.1867	-.1859	-.1828	-.1812	-.1837	-.1844	-.1883
	111	-.2188	-.2033	-.1878	-.1726	-.1618	-.1492	-.1448	-.1384	-.1379	-.1317	-.1357	-.1424
	112	-.2021	-.1831	-.1678	-.1473	-.1346	-.1176	-.1119	-.1033	-.0916	-.0777	-.0710	-.0730
	113	-.1147	-.1124	-.1051	-.0949	-.0908	-.0825	-.0765	-.0671	-.0574	-.0467	-.0362	-.0310
	114	-.1862	-.1737	-.1666	-.1503	-.1458	-.1367	-.1308	-.1264	-.1229	-.1179	-.1167	-.1176
	115	-.2281	-.2109	-.1913	-.1821	-.1691	-.1656	-.1591	-.1571	-.1531	-.1572	-.1537	-.1667
	116	-.2310	-.2224	-.2083	-.1970	-.1876	-.1801	-.1802	-.1789	-.1750	-.1756	-.1778	-.1930
	117	-.2329	-.2134	-.1988	-.1693	-.1472	-.1285	-.0974	-.0723	-.0398	-.0179	.0134	.0390
	118	-.2534	-.2294	-.2073	-.1860	-.1639	-.1480	-.1292	-.1175	-.0996	-.0842	-.0675	-.0683
	119	-.2539	-.2350	-.2127	-.1950	-.1779	-.1606	-.1473	-.1368	-.1273	-.1183	-.1101	-.1128
	120	-.2019	-.1868	-.1752	-.1622	-.1506	-.1442	-.1289	-.1297	-.1231	-.1172	-.1080	-.1176
	121	-.1539	-.1518	-.1417	-.1339	-.1311	-.1280	-.1243	-.1190	-.1210	-.1188	-.1206	-.1305
	122	-.1542	-.1302	-.1114	-.0922	-.0696	-.0412	-.0200	.0057	.0324	.0520	.0891	.1073
	123	-.1206	-.1033	-.0905	-.0719	-.0448	-.0241	-.0025	.0173	.0457	.0707	.0916	.1034
	124	-.1061	-.0845	-.0741	-.0530	-.0316	-.0179	-.0045	.0175	.0355	.0529	.0687	.0993
	125	-.0672	-.0592	-.0469	-.0302	-.0163	-.0045	.0063	.0175	.0324	.0522	.0670	.0801
	126	-.0477	-.0389	-.0300	-.0173	-.0072	.0041	.0077	.0198	.0264	.0409	.0530	.0528
Aft fuselage area-rule addition	127	.0935	.1115	.0876	.1018	.1374	.1439	.1556	.1646	.1682	.1837	.1861	.1833
	128	.0639	.0744	.0894	.1042	.1136	.1212	.1380	.1471	.1568	.1651	.1735	.1738
	129	.0578	.0566	.0507	.0521	.0492	.0543	.0561	.0609	.0655	.0680	.0734	.0728
	130	-.0782	-.0705	-.0623	-.0429	-.0344	-.0184	-.0023	.0083	.0216	.0346	.0428	.0348
	131	-.0395	-.0326	-.0384	-.0362	-.0365	-.0308	-.0280	-.0256	-.0178	-.0089	-.0077	-.0187
	132	-.0647	-.0619	-.0567	-.0486	-.0463	-.0450	-.0396	-.0324	-.0285	-.0176	-.0198	-.0282
	133	-.0810	-.0707	-.0664	-.0599	-.0541	-.0485	-.0392	-.0321	-.0263	-.0167	-.0215	-.0412
	134	-.0993	-.0916	-.0806	-.0692	-.0658	-.0570	-.0502	-.0415	-.0274	-.0193	-.0171	-.0340
	135	-.1153	-.1000	-.0884	-.0793	-.0699	-.0622	-.0525	-.0425	-.0319	-.0192	-.0165	-.0261
	136	-.1169	-.1107	-.0985	-.0850	-.0738	-.0674	-.0529	-.0450	-.0296	-.0170	-.0141	-.0156
	137	-.1287	-.1168	-.1122	-.1054	-.1084	-.1015	-.0996	-.0946	-.0881	-.0801	-.0845	-.0932
	138	-.1552	-.1433	-.1404	-.1334	-.1302	-.1223	-.1188	-.1178	-.1091	-.0972	-.1015	-.1183
	139	-.1613	-.1472	-.1390	-.1241	-.1237	-.1196	-.1068	-.1100	-.1020	-.0873	-.0918	-.1091
	140	-.1594	-.1395	-.1322	-.1249	-.1152	-.1133	-.1010	-.0974	-.0881	-.0777	-.0789	-.1037
	141	-.1616	-.1430	-.1354	-.1289	-.1190	-.1187	-.1037	-.0951	-.0908	-.0747	-.0749	-.0888
	142	-.1463	-.1378	-.1341	-.1197	-.1154	-.1089	-.0947	-.0883	-.0836	-.0705	-.0699	-.0812
	143	-.0752	-.0737	-.0796	-.0741	-.0823	-.0853	-.0847	-.0926	-.0939	-.0890	-.0981	-.0995
	144	-.0785	-.0748	-.0697	-.0727	-.0804	-.0795	-.0813	-.0819	-.0864	-.0815	-.0787	-.0895
	145	-.0500	-.0444	-.0546	-.0628	-.0661	-.0566	-.0634	-.0641	-.0675	-.0615	-.0649	-.0699
	146	-.0431	-.0512	-.0519	-.0519	-.0532	-.0522	-.0444	-.0428	-.0445	-.0428	-.0445	-.0496
	147	-.0730	-.0718	-.0714	-.0678	-.0670	-.0620	-.0623	-.0581	-.0555	-.0484	-.0513	-.0526

[redacted]

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(k) $M = 0.95$; $\beta = 0^{o}$.

	Orifice number	C_p at α, deg, of — -7.41	-6.16	-4.95	-3.66	-2.37	-1.05	0.25	1.53	2.84	4.08	5.38	6.68	7.85
Fore fuselage area-rule addition	101	.2192	.1731	.1556	.1415	.1195	.1056	.0894	.0784	.0762	.0665	.0483	.0179	-.0267
	102	.1409	.1282	.1126	.0917	.0786	.0631	.0498	.0359	.0245	.0170	.0005	-.0137	-.0229
	103	-.2049	-.2154	-.2111	-.1820	-.1602	-.1487	-.1312	-.1172	-.1028	-.0907	-.0769	-.0631	-.0393
	104	-.2011	-.1818	-.1556	-.1488	-.1065	-.0834	-.0628	-.0434	-.0248	-.0010	.0357	.0731	.1193
	105	-.1247	-.1095	-.1118	-.1174	-.1208	-.1302	-.1295	-.1348	-.1397	-.1341	-.1382	-.1421	-.1411
	106	-.1526	-.1718	-.1707	-.1629	-.1617	-.1513	-.1457	-.1378	-.1295	-.1186	-.1135	-.1136	-.1067
	107	-.1674	-.1737	-.1621	-.1379	-.1161	-.0984	-.0791	-.0575	-.0420	-.0229	-.0061	.0067	.0133
	108	-.1041	-.1197	-.1300	-.1404	-.1591	-.1676	-.1770	-.1731	-.1828	-.2175	-.2641	-.2864	-.2997
	109	-.2585	-.2375	-.2378	-.2315	-.2296	-.2257	-.2218	-.2069	-.2352	-.2529	-.2573	-.2683	-.2696
	110	-.2471	-.2287	-.2309	-.2256	-.2126	-.2051	-.1954	-.1858	-.2045	-.2136	-.2192	-.2217	-.2206
	111	-.2330	-.2276	-.2123	-.1944	-.1813	-.1688	-.1598	-.1523	-.1605	-.1630	-.1574	-.1598	-.1637
	112	-.2126	-.2019	-.1842	-.1575	-.1474	-.1326	-.1249	-.1166	-.1119	-.1043	-.0963	-.0840	-.0766
	113	-.0854	-.0941	-.0883	-.0859	-.0793	-.0682	-.0614	-.0527	-.0452	-.0308	-.0153	-.0066	.0014
	114	-.2217	-.2057	-.1828	-.1638	-.1510	-.1375	-.1281	-.1208	-.1172	-.1103	-.1055	-.1014	-.1028
	115	-.2679	-.2345	-.2103	-.1905	-.1791	-.1695	-.1622	-.1523	-.1504	-.1448	-.1479	-.1484	-.1494
	116	-.2349	-.2222	-.2033	-.1959	-.1986	-.2014	-.1775	-.1836	-.1736	-.1772	-.1800	-.1773	-.1906
	117	-.2800	-.2622	-.2458	-.2399	-.2194	-.1954	-.1659	-.1336	-.0931	-.0611	-.0304	.0034	.0294
	118	-.3257	-.3079	-.2789	-.2612	-.2351	-.2168	-.1985	-.1634	-.1380	-.1123	-.0868	-.0726	-.0597
	119	-.3407	-.3074	-.2877	-.2802	-.2546	-.2291	-.2065	-.1841	-.1598	-.1417	-.1261	-.1084	-.1033
	120	-.2672	-.2575	-.2462	-.2300	-.2194	-.2056	-.1857	-.1668	-.1548	-.1370	-.1263	-.1167	-.1181
	121	-.1967	-.2013	-.1975	-.1896	-.1826	-.1750	-.1651	-.1479	-.1445	-.1361	-.1281	-.1259	-.1246
	122	-.1382	-.1348	-.1174	-.0950	-.0789	-.0667	-.0369	-.0159	.0007	.0313	.0539	.0931	.1176
	123	-.1087	-.1062	-.0861	-.0686	-.0514	-.0328	-.0163	.0055	.0252	.0488	.0789	.0952	.1161
	124	-.0912	-.0853	-.0701	-.0544	-.0427	-.0224	-.0090	.0041	.0259	.0488	.0614	.0821	.1017
	125	-.0560	-.0510	-.0447	-.0301	-.0291	-.0092	.0038	.0153	.0264	.0418	.0576	.0770	.0935
	126	-.0396	-.0378	-.0265	-.0153	-.0132	.0039	.0101	.0150	.0247	.0371	.0497	.0641	.0729
Aft fuselage area-rule addition	127	.0610	.0625	.0785	.0979	.1192	.1453	.1497	.1552	.1569	.1631	.1749	.1808	.1801
	128	-.0023	.0173	.0363	.0762	.1051	.1239	.1338	.1411	.1442	.1485	.1588	.1672	.1714
	129	.0420	.0518	.0513	.0573	.0580	.0641	.0619	.0638	.0527	.0399	.0294	.0171	.0109
	130	-.1244	-.0904	-.0763	-.0602	-.0447	-.0238	-.0103	-.0021	.0033	.0081	.0126	.0169	.0241
	131	-.0427	-.0412	-.0430	-.0347	-.0376	-.0350	-.0318	-.0297	-.0257	-.0308	-.0449	-.0522	-.0590
	132	-.0866	-.0713	-.0637	-.0546	-.0541	-.0437	-.0454	-.0400	-.0328	-.0400	-.0507	-.0665	-.0691
	133	-.0987	-.0884	-.0742	-.0659	-.0635	-.0504	-.0458	-.0413	-.0354	-.0376	-.0456	-.0537	-.0624
	134	-.1215	-.1037	-.0953	-.0783	-.0718	-.0604	-.0584	-.0483	-.0403	-.0418	-.0452	-.0546	-.0612
	135	-.1217	-.1114	-.1027	-.0878	-.0851	-.0711	-.0626	-.0519	-.0459	-.0437	-.0459	-.0490	-.0498
	136	-.1343	-.1291	-.1091	-.1001	-.0946	-.0761	-.0657	-.0575	-.0451	-.0454	-.0447	-.0466	-.0447
	137	-.1523	-.1432	-.1290	-.1219	-.1191	-.1089	-.1088	-.1081	-.0944	-.0839	-.0762	-.0745	-.0791
	138	-.1789	-.1727	-.1589	-.1503	-.1466	-.1362	-.1367	-.1317	-.1164	-.1065	-.1057	-.1055	-.1077
	139	-.1916	-.1751	-.1619	-.1512	-.1437	-.1338	-.1261	-.1227	-.1103	-.0972	-.0888	-.0885	-.0963
	140	-.1998	-.1776	-.1635	-.1520	-.1386	-.1292	-.1215	-.1159	-.0982	-.0898	-.0847	-.0868	-.0888
	141	-.1966	-.1761	-.1635	-.1537	-.1443	-.1341	-.1263	-.1171	-.0992	-.0895	-.0847	-.0834	-.0832
	142	-.1836	-.1636	-.1520	-.1452	-.1392	-.1297	-.1214	-.1132	-.0980	-.0856	-.0822	-.0811	-.0853
	143	-.0869	-.0640	-.0711	-.0844	-.0885	-.0892	-.0917	-.1035	-.0992	-.0919	-.0915	-.0925	-.0946
	144	-.0781	-.0563	-.0640	-.0711	-.0824	-.0817	-.0916	-.0870	-.0919	-.0825	-.0793	-.0813	-.0839
	145	-.0476	-.0390	-.0430	-.0521	-.0716	-.0681	-.0708	-.0791	-.0752	-.0638	-.0606	-.0563	-.0635
	146	-.0304	-.0231	-.0332	-.0483	-.0521	-.0556	-.0589	-.0563	-.0568	-.0493	-.0456	-.0464	-.0461
	147	-.0636	-.0604	-.0601	-.0633	-.0720	-.0662	-.0711	-.0692	-.0691	-.0573	-.0541	-.0505	-.0449

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(l) M = 0.98; $\beta = 0^{\circ}$.

	Orifice number	C_p at α, deg, of —														
		-7.36	-6.12	-4.90	-3.65	-2.39	-1.09	0.23	1.57	2.88	4.12	5.42	6.61	7.84	9.00	10.19
Fore fuselage area-rule addition	101	.2192	.1987	.1778	.1500	.1374	.1182	.1039	.0984	.0940	.0791	.0600	.0316	-.0154	-.0633	-.1262
	102	.1671	.1512	.1237	.1053	.0915	.0801	.0697	.0580	.0527	.0355	.0207	.0076	-.0011	-.0069	-.0012
	103	-.2125	-.1961	-.1889	-.1784	-.1613	-.1291	-.1166	-.0977	-.0779	-.0701	-.0522	-.0403	-.0159	.0125	.0503
	104	-.1505	-.1362	-.1372	-.1241	-.1066	-.0894	-.0433	-.0217	.0035	.0219	.0548	.0920	.1434	.1683	.2087
	105	-.1516	-.1528	-.1614	-.1069	-.1173	-.1224	-.1234	-.1201	-.1225	-.1239	-.1248	-.1243	-.1216	-.1201	-.1129
	106	-.2055	-.2902	-.2264	-.1335	-.1463	-.1415	-.1342	-.1224	-.1120	-.1058	-.1011	-.0950	-.0904	-.0834	-.0894
	107	-.3896	-.1956	-.0798	-.1087	-.1042	-.0810	-.0667	-.0442	-.0220	-.0094	.0054	.0177	.0318	.0369	.0465
	108	-.0661	-.1033	-.1321	-.1440	-.1935	-.2116	-.2269	-.2398	-.2549	-.2653	-.2845	-.2959	-.3053	-.3146	-.3296
	109	-.1758	-.2170	-.2315	-.2401	-.2567	-.2528	-.2555	-.2537	-.2534	-.2534	-.2521	-.2564	-.2593	-.2604	-.2644
	110	-.1787	-.2100	-.2151	-.2251	-.2282	-.2264	-.2237	-.2164	-.2091	-.2074	-.2090	-.2087	-.2080	-.2181	-.2188
	111	-.1867	-.1927	-.1969	-.2001	-.1981	-.1903	-.1817	-.1687	-.1619	-.1580	-.1533	-.1493	-.1517	-.1576	-.1614
	112	-.1672	-.1683	-.1519	-.1662	-.1594	-.1458	-.1323	-.1183	-.1063	-.0992	-.0875	-.0735	-.0646	-.0558	-.0451
	113	-.0593	-.0726	-.0771	-.0764	-.0595	-.0481	-.0416	-.0351	-.0321	-.0246	-.0135	.0068	.0226	.0444	.0482
	114	-.2140	-.2094	-.1867	-.1682	-.1522	-.1344	-.1207	-.1217	-.1357	-.1353	-.1316	-.1192	-.1030	-.0991	-.1001
	115	-.2578	-.2419	-.2274	-.2055	-.1933	-.1789	-.1783	-.2140	-.2663	-.2984	-.3135	-.3182	-.3415	-.3587	-.3874
	116	-.2692	-.2542	-.2441	-.2245	-.2144	-.2072	-.2288	-.2849	-.2980	-.3016	-.3093	-.3128	-.3224	-.3209	-.3174
	117	-.2770	-.2719	-.2489	-.2275	-.2132	-.1736	-.1536	-.1271	-.0912	-.0590	-.0125	.0272	.0525	.0760	.1055
	118	-.3314	-.3064	-.2872	-.2659	-.2395	-.2113	-.1814	-.1616	-.1336	-.1086	-.0669	-.0394	-.0273	-.0130	-.0004
	119	-.3387	-.3220	-.2991	-.2752	-.2424	-.2228	-.1988	-.1793	-.1517	-.1385	-.0974	-.0749	-.0684	-.0619	-.0581
	120	-.2755	-.2557	-.2455	-.2292	-.2109	-.1910	-.1792	-.1651	-.1507	-.1331	-.0974	-.0759	-.0786	-.0788	-.0785
	121	-.1970	-.1944	-.1860	-.1762	-.1628	-.1545	-.1521	-.1445	-.1404	-.1326	-.1055	-.0829	-.0836	-.0887	-.0870
	122	-.1171	-.0982	-.0771	-.0663	-.0476	-.0461	-.0094	.0097	.0318	.0450	.0527	.0871	.1228	.1502	.1735
	123	-.0779	-.0614	-.0488	-.0450	-.0259	-.0087	.0131	.0360	.0542	.0714	.0881	.1140	.1347	.1501	.1684
	124	-.0542	-.0512	-.0334	-.0273	-.0109	-.0004	.0182	.0382	.0530	.0688	.0820	.1010	.1257	.1450	.1664
	125	-.0240	-.0116	-.0079	-.0006	.0091	.0151	.0284	.0445	.0552	.0629	.0815	.0939	.1104	.1283	.1477
	126	-.0075	-.0043	.0051	.0108	.0216	.0267	.0352	.0427	.0525	.0609	.0702	.0830	.0908	.1051	.1178
Aft fuselage area-rule addition	127	.0610	.0767	.0389	.0992	.1274	.1505	.1708	.1731	.1752	.1769	.1915	.2007	.2006	.2017	.2069
	128	.0172	.0339	.0554	.0803	.1042	.1335	.1560	.1560	.1623	.1632	.1784	.1875	.1937	.1908	.1901
	129	.0514	.0549	.0552	.0576	.0648	.0793	.0824	.0751	.0674	.0552	.0462	.0400	.0358	.0213	.0167
	130	-.1370	-.1126	-.0977	-.0737	-.0367	-.0130	.0036	.0111	.0129	.0170	.0256	.0316	.0411	.0468	.0569
	131	-.0563	-.0456	-.0488	-.0484	-.0367	-.0248	-.0242	-.0191	-.0384	-.0515	-.0556	-.0628	-.0694	-.0820	-.0753
	132	-.0982	-.0785	-.0801	-.0747	-.0546	-.0372	-.0329	-.0302	-.0472	-.0569	-.0635	-.0698	-.0791	-.0827	-.0712
	133	-.1116	-.0957	-.0955	-.0839	-.0634	-.0427	-.0381	-.0309	-.0450	-.0558	-.0583	-.0614	-.0733	-.0868	-.0780
	134	-.1321	-.1136	-.1117	-.1002	-.0750	-.0548	-.0474	-.0394	-.0510	-.0546	-.0549	-.0588	-.0633	-.0812	-.0855
	135	-.1440	-.1250	-.1265	-.1129	-.0863	-.0614	-.0537	-.0420	-.0535	-.0554	-.0512	-.0541	-.0565	-.0621	-.0574
	136	-.1604	-.1369	-.1357	-.1224	-.0940	-.0706	-.0606	-.0483	-.0515	-.0525	-.0486	-.0469	-.0479	-.0514	-.0407
	137	-.1692	-.1778	-.1386	-.1152	-.1122	-.1126	-.1069	-.1065	-.1098	-.1295	-.1310	-.1359	-.1448	-.1511	-.1534
	138	-.2002	-.2075	-.1645	-.1497	-.1436	-.1407	-.1330	-.1338	-.1285	-.1527	-.1618	-.1678	-.1780	-.1823	-.1941
	139	-.2099	-.2138	-.1739	-.1457	-.1391	-.1402	-.1269	-.1236	-.1307	-.1461	-.1461	-.1475	-.1628	-.1770	-.1928
	140	-.2176	-.2259	-.1846	-.1636	-.1475	-.1381	-.1212	-.1212	-.1237	-.1329	-.1363	-.1381	-.1470	-.1581	-.1681
	141	-.2244	-.2331	-.1741	-.1505	-.1413	-.1412	-.1262	-.1244	-.1193	-.1357	-.1325	-.1373	-.1415	-.1509	-.1638
	142	-.2127	-.2319	-.1756	-.1294	-.1465	-.1383	-.1247	-.1179	-.1176	-.1278	-.1284	-.1294	-.1380	-.1465	-.1516
	143	-.3033	-.2829	-.2755	-.2789	-.2760	-.2751	-.2689	-.2531	-.2113	-.1568	-.1392	-.1782	-.1683	-.1342	-.3006
	144	-.2801	-.2319	-.2748	-.2666	-.2533	-.2456	-.2396	-.2258	-.1619	-.1220	-.1281	-.1627	-.1298	-.1417	-.2866
	145	-.2070	-.1010	-.2068	-.2085	-.1993	-.1936	-.2022	-.1706	-.1255	-.0881	-.0817	-.1197	-.0991	-.1170	-.2190
	146	-.1215	-.0657	-.1632	-.1701	-.1702	-.1583	-.1596	-.1347	-.0955	-.0679	-.0918	-.1008	-.0955	-.1006	-.2067
	147	-.1535	-.0713	-.1615	-.1708	-.1516	-.1489	-.1489	-.1335	-.0955	-.0743	-.0924	-.1202	-.1201	-.1075	-.1911

[redacted]

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(m) M = 0.99; $\beta = 0^{o}$.

	Orifice number	C_p at α, deg, of —														
		-7.37	-6.12	-4.91	-3.62	-2.39	-1.13	0.19	1.53	2.84	4.11	5.37	6.62	7.83	8.99	10.18
Fore fuselage area-rule addition	101	.2251	.2038	.1846	.1623	.1374	.1313	.1151	.1094	.1084	.0955	.0675	.0354	-.0097	-.0529	-.1209
	102	.1708	.1579	.1359	.1193	.1007	.0869	.0811	.0710	.0615	.0500	.0330	.0201	.0105	.0026	.0049
	103	-.1864	-.1780	-.1822	-.1560	-.1497	-.1339	-.1087	-.0821	-.0675	-.0496	-.0421	-.0302	-.0128	.0281	.0568
	104	-.1508	-.1307	-.1170	-.1108	-.0997	-.0722	-.0346	-.0069	.0136	.0318	.0623	.0982	.1430	.1736	.1953
	105	-.1387	-.1430	-.1522	-.1441	-.1024	-.1034	-.1075	-.1110	-.1097	-.1103	-.1143	-.1131	-.1112	-.1109	-.1078
	106	-.2964	-.2746	-.2560	-.1089	-.1276	-.1207	-.1182	-.1115	-.1017	-.0910	-.0872	-.0872	-.0795	-.0767	-.0838
	107	-.3975	-.2978	-.1618	-.0708	-.0849	-.0707	-.0533	-.0373	-.0095	.0046	.0154	.0311	.0388	.0458	.0500
	108	-.1673	-.1074	-.1070	-.1651	-.1895	-.2036	-.2159	-.2289	-.2426	-.2563	-.2741	-.2874	-.2914	-.3104	-.3233
	109	-.2575	-.1719	-.1983	-.2401	-.2464	-.2455	-.2412	-.2415	-.2395	-.2372	-.2402	-.2400	-.2476	-.2536	-.2598
	110	-.2325	-.1804	-.1973	-.2216	-.2190	-.2167	-.2069	-.2048	-.1956	-.1928	-.1953	-.1961	-.1971	-.2074	-.2145
	111	-.2337	-.1818	-.1839	-.1947	-.1885	-.1796	-.1686	-.1585	-.1502	-.1474	-.1412	-.1391	-.1416	-.1497	-.1563
	112	-.1889	-.1089	-.1497	-.1588	-.1482	-.1349	-.1180	-.1042	-.0922	-.0860	-.0736	-.0629	-.0535	-.0479	-.0429
	113	-.0309	-.0493	-.0581	-.0628	-.0536	-.0493	-.0535	-.0557	-.0546	-.0552	-.0535	-.0459	-.0293	-.0037	.0207
	114	-.1709	-.1876	-.1682	-.1577	-.1530	-.1439	-.1502	-.1593	-.1755	-.1928	-.2220	-.2505	-.2582	-.2483	-.2258
	115	-.2208	-.2228	-.2091	-.2015	-.1878	-.1968	-.2205	-.2574	-.2795	-.2923	-.3126	-.3191	-.3341	-.3569	-.3863
	116	-.2405	-.2397	-.2257	-.2156	-.2113	-.2433	-.2749	-.2849	-.2914	-.2906	-.3000	-.3066	-.3077	-.3159	-.3141
	117	-.2628	-.2448	-.2298	-.2166	-.1919	-.1640	-.1470	-.1207	-.0881	-.0682	-.0382	.0095	.0535	.0876	.1147
	118	-.3132	-.2937	-.2703	-.2423	-.2208	-.2018	-.1778	-.1517	-.1379	-.1219	-.0995	-.0695	-.0188	.0087	.0190
	119	-.3210	-.3063	-.2875	-.2616	-.2319	-.2093	-.1868	-.1757	-.1595	-.1503	-.1427	-.1185	-.0645	-.0333	-.0264
	120	-.2647	-.2504	-.2339	-.2166	-.1923	-.1807	-.1660	-.1602	-.1529	-.1503	-.1466	-.1410	-.0855	-.0525	-.0419
	121	-.1923	-.1847	-.1771	-.1663	-.1606	-.1473	-.1449	-.1471	-.1451	-.1552	-.1681	-.1886	-.1850	-.1273	-.0655
	122	-.0989	-.1080	-.1043	-.0949	-.0694	-.0450	-.0242	-.0026	.0146	.0501	.0551	.0858	.1323	.1578	.1678
	123	-.0635	-.0668	-.0695	-.0609	-.0390	-.0149	.0055	.0278	.0562	.0797	.1051	.1308	.1466	.1535	.1678
	124	-.0354	-.0501	-.0510	-.0415	-.0267	-.0076	.0142	.0348	.0567	.0790	.1037	.1192	.1391	.1435	.1707
	125	-.0102	-.0179	-.0214	-.0142	-.0044	.0084	.0305	.0465	.0644	.0782	.0939	.1131	.1286	.1408	.1537
	126	.0004	.0011	-.0085	-.0051	.0071	.0245	.0368	.0479	.0522	.0732	.0898	.1012	.1107	.1122	.1248
Aft fuselage area-rule addition	127	.0679	.0813	.0940	.1112	.1274	.1392	.1602	.1691	.1810	.1815	.1925	.2023	.2055	.2093	.2331
	128	.0241	.0417	.0528	.0851	.1042	.1203	.1480	.1538	.1618	.1707	.1842	.1959	.2024	.2004	.2065
	129	.0641	.0663	.0617	.0696	.0781	.0762	.0920	.0760	.0651	.0569	.0539	.0498	.0436	.0373	.0330
	130	-.1232	-.1000	-.0873	-.0633	-.0301	-.0040	.0188	.0143	.0137	.0220	.0338	.0431	.0530	.0578	.0713
	131	-.0466	-.0404	-.0478	-.0394	-.0264	-.0149	-.0054	-.0143	-.0272	-.0357	-.0386	-.0489	-.0554	-.0653	-.0770
	132	-.0894	-.0731	-.0787	-.0628	-.0466	-.0255	-.0144	-.0242	-.0350	-.0399	-.0454	-.0522	-.0572	-.0685	-.0620
	133	-.1042	-.0851	-.0915	-.0769	-.0509	-.0384	-.0196	-.0246	-.0360	-.0391	-.0467	-.0523	-.0622	-.0745	-.0700
	134	-.1270	-.1019	-.1131	-.0944	-.0633	-.0448	-.0302	-.0358	-.0386	-.0445	-.0423	-.0459	-.0489	-.0718	-.0748
	135	-.1379	-.1155	-.1216	-.1033	-.0764	-.0545	-.0322	-.0377	-.0464	-.0438	-.0421	-.0401	-.0443	-.0495	-.0450
	136	-.1535	-.1317	-.1378	-.1227	-.0852	-.0624	-.0384	-.0429	-.0473	-.0433	-.0375	-.0321	-.0334	-.0376	-.0291
	137	-.1644	-.1740	-.1443	-.1263	-.1062	-.0995	-.0915	-.0933	-.1024	-.1129	-.1189	-.1269	-.1352	-.1440	-.1488
	138	-.1993	-.1956	-.1710	-.1513	-.1365	-.1272	-.1162	-.1192	-.1293	-.1398	-.1477	-.1532	-.1655	-.1761	-.1812
	139	-.2063	-.2031	-.1800	-.1602	-.1356	-.1254	-.1158	-.1091	-.1233	-.1294	-.1363	-.1423	-.1532	-.1715	-.1800
	140	-.2165	-.2201	-.1880	-.1595	-.1343	-.1260	-.1105	-.1052	-.1153	-.1182	-.1230	-.1277	-.1406	-.1530	-.1556
	141	-.2256	-.2322	-.1967	-.1712	-.1465	-.1269	-.1170	-.1112	-.1169	-.1197	-.1197	-.1250	-.1326	-.1429	-.1509
	142	-.2160	-.2225	-.1912	-.1649	-.1382	-.1240	-.1111	-.1064	-.1131	-.1153	-.1150	-.1190	-.1238	-.1370	-.1364
	143	-.3209	-.3147	-.3033	-.2860	-.2677	-.2660	-.2564	-.2502	-.2598	-.2643	-.2699	-.2732	-.2739	-.2817	-.3035
	144	-.3269	-.3217	-.3054	-.2858	-.2595	-.2547	-.2433	-.2359	-.2392	-.2429	-.2503	-.2562	-.2587	-.2613	-.2840
	145	-.2986	-.2976	-.2768	-.2521	-.2357	-.2321	-.2178	-.2087	-.2080	-.2133	-.2190	-.2223	-.2239	-.2362	-.2536
	146	-.2365	-.2287	-.2224	-.2099	-.1933	-.1900	-.1803	-.1708	-.1696	-.1757	-.1911	-.1966	-.1937	-.1981	-.2277
	147	-.2264	-.2319	-.2118	-.1978	-.1824	-.1798	-.1718	-.1607	-.1677	-.1726	-.1836	-.1905	-.1879	-.1930	-.2107

[redacted]

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(n) M = 1.00; $\beta = 0^{o}$.

	Orifice number	C_p at α, deg, of —														
		-7.34	-6.10	-4.88	-3.65	-2.39	-1.09	0.16	1.51	2.84	4.12	5.35	6.61	7.84	9.02	10.20
Fore fuselage area-rule addition	101	.2351	.2172	.1944	.1780	.1552	.1346	.1199	.1163	.1084	.1005	.0755	.0435	-.0004	-.0523	-.1209
	102	.1903	.1682	.1446	.1302	.1082	.0989	.0907	.0824	.0720	.0566	.0471	.0301	.0185	.0139	.0207
	103	-.1815	-.1714	-.1633	-.1505	-.1420	-.1207	-.1055	-.0730	-.0612	-.0465	-.0353	-.0157	-.0051	.0383	.0684
	104	-.1264	-.1243	-.1002	-.0996	-.0841	-.0519	-.0321	.0008	.0212	.0430	.0716	.1135	.1547	.1808	.2275
	105	-.1251	-.1313	-.1377	-.1459	-.0871	-.0999	-.0995	-.0964	-.1013	-.1014	-.1004	-.1028	-.1001	-.1016	-.0989
	106	-.2812	-.2634	-.2480	-.2083	-.1042	-.1127	-.1072	-.0996	-.0938	-.0842	-.0770	-.0763	-.0691	-.0653	-.0740
	107	-.3882	-.3277	-.1650	-.0871	-.0720	-.0539	-.0428	-.0261	-.0055	.0110	.0257	.0360	.0463	.0532	.0599
	108	-.1912	-.2009	-.1554	-.1545	-.1809	-.1949	-.2110	-.2239	-.2358	-.2490	-.2611	-.2728	-.2892	-.2991	-.3135
	109	-.3442	-.2486	-.2388	-.2324	-.2343	-.2371	-.2345	-.2324	-.2330	-.2285	-.2277	-.2319	-.2351	-.2423	-.2496
	110	-.2563	-.2316	-.2222	-.2075	-.2107	-.2039	-.2016	-.1948	-.1909	-.1822	-.1812	-.1845	-.1855	-.1968	-.2013
	111	-.2452	-.2251	-.2066	-.1841	-.1784	-.1596	-.1631	-.1504	-.1461	-.1382	-.1292	-.1268	-.1313	-.1369	-.1435
	112	-.2108	-.1676	-.1713	-.1471	-.1402	-.1265	-.1128	-.0979	-.0892	-.0787	-.0620	-.0437	-.0420	-.0373	-.0287
	113	-.0366	-.0464	-.0484	-.0513	-.0464	-.0500	-.0559	-.0577	-.0638	-.0598	-.0573	-.0573	-.0534	-.0479	-.0350
	114	-.1943	-.1859	-.1687	-.1545	-.1572	-.1539	-.1599	-.1678	-.1863	-.2192	-.2403	-.2617	-.2858	-.3059	-.3225
	115	-.2336	-.2128	-.2040	-.1886	-.1951	-.2133	-.2390	-.2609	-.2794	-.2884	-.3011	-.3088	-.3287	-.3497	-.3718
	116	-.2386	-.2179	-.2107	-.2000	-.2232	-.2576	-.2715	-.2766	-.2867	-.2836	-.2875	-.2958	-.3003	-.3027	-.3062
	117	-.2795	-.2520	-.2202	-.2039	-.1836	-.1691	-.1393	-.1153	-.0868	-.0631	-.0416	-.0210	.0252	.0748	.1053
	118	-.3202	-.2781	-.2660	-.2327	-.2068	-.1839	-.1681	-.1475	-.1330	-.1181	-.1052	-.0886	-.0674	-.0244	.0269
	119	-.3168	-.2653	-.2560	-.2325	-.2143	-.1937	-.1808	-.1634	-.1509	-.1483	-.1389	-.1360	-.1229	-.0890	-.0239
	120	-.2542	-.2333	-.2101	-.2051	-.1862	-.1710	-.1553	-.1509	-.1492	-.1474	-.1461	-.1515	-.1504	-.1455	-.0748
	121	-.1781	-.1608	-.1554	-.1540	-.1478	-.1437	-.1403	-.1402	-.1487	-.1573	-.1703	-.2031	-.2430	-.2735	-.2981
	122	-.0833	-.1144	-.1094	-.0847	-.0704	-.0469	-.0462	-.0145	.0030	.0235	.0560	.0811	.1328	.1720	.1758
	123	-.0661	-.0853	-.0803	-.0662	-.0479	-.0242	-.0101	.0141	.0318	.0619	.0959	.1318	.1564	.1662	.1739
	124	-.0455	-.0668	-.0585	-.0466	-.0287	-.0172	-.0007	.0167	.0355	.0597	.0888	.1225	.1498	.1597	.1759
	125	-.0182	-.0356	-.0321	-.0273	-.0097	-.0019	.0124	.0299	.0442	.0621	.0910	.1172	.1396	.1522	.1700
	126	-.0029	-.0179	-.0198	-.0096	-.0021	.0098	.0185	.0308	.0403	.0587	.0832	.1145	.1215	.1347	.1416
Aft fuselage area-rule addition	127	.0798	.0979	.1064	.1163	.1226	.1388	.1395	.1703	.1310	.1912	.2003	.2096	.2123	.2213	.2331
	128	.0366	.0569	.0670	.0891	.1015	.1171	.1276	.1540	.1593	.1778	.1936	.2020	.2082	.2053	.2085
	129	.0749	.0805	.0763	.0776	.0755	.0798	.0753	.0731	.0636	.0560	.0538	.0509	.0514	.0436	.0378
	130	-.1183	-.0931	-.0765	-.0435	-.0278	-.0090	-.0012	.0154	.0227	.0273	.0375	.0522	.0593	.0686	.0812
	131	-.0397	-.0258	-.0363	-.0283	-.0218	-.0234	-.0143	-.0138	-.0324	-.0345	-.0370	-.0375	-.0428	-.0552	-.0563
	132	-.0815	-.0650	-.0685	-.0488	-.0379	-.0414	-.0280	-.0236	-.0384	-.0431	-.0436	-.0445	-.0498	-.0523	-.0522
	133	-.1006	-.0796	-.0820	-.0619	-.0522	-.0408	-.0247	-.0268	-.0380	-.0448	-.0402	-.0438	-.0520	-.0586	-.0553
	134	-.1127	-.0969	-.1030	-.0721	-.0616	-.0532	-.0326	-.0363	-.0450	-.0460	-.0404	-.0360	-.0481	-.0617	-.0675
	135	-.1314	-.1049	-.1139	-.0865	-.0706	-.0634	-.0428	-.0369	-.0434	-.0428	-.0402	-.0355	-.0326	-.0425	-.0392
	136	-.1536	-.1194	-.1299	-.1003	-.0807	-.0691	-.0460	-.0420	-.0513	-.0465	-.0356	-.0249	-.0256	-.0269	-.0208
	137	-.1599	-.1579	-.1343	-.1127	-.1004	-.0849	-.0888	-.0884	-.0963	-.1059	-.1055	-.1089	-.1194	-.1309	-.1405
	138	-.1861	-.1816	-.1553	-.1402	-.1346	-.1129	-.1127	-.1105	-.1219	-.1314	-.1369	-.1377	-.1535	-.1589	-.1725
	139	-.1957	-.1895	-.1681	-.1451	-.1294	-.1093	-.1055	-.1059	-.1156	-.1202	-.1231	-.1304	-.1458	-.1553	-.1724
	140	-.2028	-.1997	-.1728	-.1526	-.1352	-.1108	-.1033	-.1017	-.1083	-.1108	-.1113	-.1135	-.1265	-.1386	-.1468
	141	-.2224	-.2142	-.1803	-.1535	-.1349	-.1148	-.1086	-.1070	-.1117	-.1110	-.1122	-.1161	-.1194	-.1280	-.1407
	142	-.2098	-.2053	-.1790	-.1605	-.1361	-.1129	-.1045	-.1020	-.1066	-.1071	-.1062	-.1077	-.1168	-.1202	-.1332
	143	-.3124	-.3058	-.2891	-.2815	-.2592	-.2433	-.2445	-.2366	-.2518	-.2613	-.2594	-.2652	-.2667	-.2715	-.2878
	144	-.3233	-.3087	-.2983	-.2839	-.2563	-.2346	-.2286	-.2279	-.2378	-.2383	-.2408	-.2447	-.2488	-.2512	-.2718
	145	-.3039	-.2906	-.2732	-.2558	-.2317	-.2176	-.2078	-.1984	-.2129	-.2184	-.2091	-.2158	-.2197	-.2289	-.2399
	146	-.2423	-.2358	-.2255	-.2179	-.1927	-.1720	-.1729	-.1651	-.1751	-.1752	-.1826	-.1858	-.1878	-.1906	-.2126
	147	-.2311	-.2404	-.2168	-.2012	-.1767	-.1645	-.1630	-.1597	-.1708	-.1749	-.1774	-.1800	-.1827	-.1875	-.2095

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(o) M = 0.50; β = 5.3°

	Orifice number	C_p at α, deg, of –										
		-6.61	-4.37	-2.14	0.10	2.35	4.55	6.75	8.96	11.14	13.34	14.38
Fore fuselage area-rule addition	101	-.0096	-.0549	-.0461	-.0636	-.0984	-.1517	-.2427	-.4071	-.4812	-.4913	-.4992
	102	.0198	.0029	-.0200	-.0451	-.0740	-.0940	-.1287	-.1418	-.1528	-.1713	-.1836
	103	-.2997	-.2685	-.2475	-.2259	-.2020	-.1903	-.1453	-.0743	-.0875	-.1452	-.1654
	104	-.2711	-.2235	-.1818	-.1468	-.1104	-.0601	.0014	.0051	.0305	.0719	.0713
	105	-.1581	-.1480	-.1494	-.1502	-.1534	-.1486	-.1626	-.1751	-.1826	-.2005	-.2087
	106	-.2523	-.2073	-.1887	-.1656	-.1549	-.1420	-.1393	-.1487	-.1751	-.2096	-.2219
	107	-.2013	-.1493	-.1227	-.0940	-.0680	-.0507	-.0542	-.0636	-.0831	-.1016	-.1289
	108	-.1653	-.1602	-.1658	-.1681	-.1766	-.1728	-.1773	-.1870	-.2005	-.2096	-.2181
	109	-.2736	-.2409	-.2201	-.1964	-.1885	-.1731	-.1824	-.1927	-.2052	-.2415	-.2590
	110	-.2633	-.2277	-.2051	-.1870	-.1810	-.1838	-.1833	-.2052	-.2316	-.2633	-.2885
	111	-.2360	-.1957	-.1727	-.1543	-.1440	-.1383	-.1550	-.1829	-.2061	-.2398	-.2665
	112	-.1665	-.1417	-.1290	-.1167	-.1053	-.1119	-.1158	-.1280	-.1455	-.1757	-.2005
	113	-.1709	-.1483	-.1381	-.1267	-.1151	-.1034	-.0957	-.0843	-.0912	-.1054	-.1126
	114	-.2436	-.2142	-.1856	-.1659	-.1527	-.1364	-.1374	-.1478	-.1594	-.1672	-.1820
	115	-.2480	-.2067	-.1871	-.1725	-.1597	-.1674	-.1651	-.1836	-.2055	-.2438	-.2668
	116	-.2032	-.1791	-.1705	-.1703	-.1735	-.1778	-.1968	-.2281	-.2670	-.3110	-.3415
	117	-.2689	-.2331	-.1887	-.1393	-.1044	-.0673	-.0325	-.0153	-.0131	.0130	.0064
	118	-.2555	-.2020	-.1721	-.1399	-.1173	-.1000	-.0859	-.0934	-.1041	-.1041	-.1139
	119	-.2360	-.1847	-.1598	-.1418	-.1314	-.1213	-.1302	-.1339	-.1534	-.1754	-.1858
	120	-.1718	-.1455	-.1337	-.1295	-.1239	-.1169	-.1312	-.1556	-.1788	-.2112	-.2326
	121	-.1258	-.1169	-.1127	-.1179	-.1214	-.1354	-.1651	-.1917	-.2426	-.2749	-.2901
	122	-.1675	-.1389	-.1108	-.0655	-.0247	.0133	.0591	.0788	.0860	.0653	.0462
	123	-.1584	-.1198	-.0847	-.0426	-.0099	.0145	.0409	.0609	.0575	.0634	.0584
	124	-.1280	-.0953	-.0658	-.0366	-.0152	-.0030	.0356	.0371	.0628	.0591	.0640
	125	-.0932	-.0717	-.0426	-.0284	-.0184	-.0011	.0127	.0183	.0258	.0405	.0409
	126	-.0744	-.0482	-.0417	-.0300	-.0171	-.0030	-.0127	-.0209	-.0247	-.0316	-.0394
Aft fuselage area-rule addition	127	.0469	.0718	.0752	.0970	.1198	.1317	.1455	.1464	.1765	.1658	.1458
	128	.0282	.0397	.0568	.0748	.0849	.1064	.0986	.0625	.0181	-.0165	-.0181
	129	.0143	.0162	.0128	.0197	.0349	.0339	.0463	.0492	.0568	.0445	.0359
	130	-.0616	-.0491	-.0357	-.0262	-.0174	-.0115	-.0118	-.0011	.0115	.0087	.0102
	131	-.0362	-.0378	-.0435	-.0388	-.0335	-.0218	-.0171	-.0178	-.0162	-.0366	-.0303
	132	-.0528	-.0473	-.0473	-.0416	-.0363	-.0294	-.0193	-.0124	-.0121	-.0297	-.0234
	133	-.0613	-.0520	-.0530	-.0426	-.0347	-.0272	-.0131	-.0131	-.0074	-.0416	-.0721
	134	-.0685	-.0648	-.0583	-.0482	-.0444	-.0338	-.0413	-.0727	-.1151	-.1587	-.1735
	135	-.0788	-.0648	-.0633	-.0526	-.0422	-.0400	-.0599	-.1176	-.1132	-.0959	-.0592
	136	-.0772	-.0683	-.0652	-.0520	-.0466	-.0482	-.0536	-.0401	-.0492	-.0495	-.0517
	137	-.0998	-.0909	-.0979	-.0843	-.0827	-.0767	-.0677	-.0790	-.0787	-.0997	-.1004
	138	-.1064	-.1078	-.0979	-.0950	-.0909	-.0874	-.0825	-.0859	-.0940	-.1135	-.1230
	139	-.1001	-.0934	-.0922	-.0818	-.0946	-.0771	-.0642	-.0796	-.0878	-.1361	-.1393
	140	-.0882	-.0915	-.0850	-.0902	-.0736	-.0705	-.0740	-.1088	-.1386	-.1292	-.1019
	141	-.0916	-.0818	-.0850	-.0802	-.0689	-.0783	-.0950	-.1192	-.0696	-.0708	-.0727
	142	-.0832	-.0771	-.0750	-.0718	-.0708	-.0749	-.0740	-.0674	-.0740	-.0736	-.0693
	143	-.0600	-.0749	-.0812	-.0881	-.0937	-.1006	-.1023	-.1236	-.1286	-.1490	-.1553
	144	-.0541	-.0586	-.0636	-.0696	-.0714	-.0783	-.0875	-.0963	-.1110	-.1283	-.1277
	145	-.0506	-.0407	-.0530	-.0532	-.0570	-.0626	-.0621	-.0821	-.0925	-.0988	-.0891
	146	-.0359	-.0356	-.0373	-.0347	-.0394	-.0407	-.0526	-.0724	-.0517	-.0426	-.0460
	147	-.0613	-.0513	-.0489	-.0419	-.0457	-.0538	-.0586	-.0413	-.0391	-.0536	-.0397

TABLE X - Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(p) M = 0.80; β = 5.8°.

	Orifice number	C_p at α, deg, of —							
		-7.46	-4.97	-2.35	0.26	2.87	5.53	7.99	10.31
Fore fuselage area-rule addition	101	-.0212	-.0549	-.0891	-.1792	-.1908	-.3063	-.4145	-.4684
	102	.0109	-.0161	-.0388	-.0700	-.0906	-.1236	-.1521	-.1482
	103	-.4215	-.3523	-.3125	-.2746	-.2423	-.1946	-.1157	-.1631
	104	-.3537	-.2897	-.2236	-.1585	-.1023	-.0423	-.0911	-.1704
	105	-.1926	-.1859	-.1820	-.1799	-.1747	-.1727	-.1882	-.1869
	106	-.2795	-.2425	-.2140	-.1831	-.1549	-.1559	-.1480	-.1548
	107	-.2148	-.1749	-.1301	-.0968	-.0665	-.0503	-.0483	-.0606
	108	-.1707	-.1807	-.1846	-.1929	-.1955	-.2011	-.1960	-.2017
	109	-.3087	-.2710	-.2449	-.2223	-.2053	-.2044	-.2027	-.2167
	110	-.2901	-.2502	-.2269	-.2069	-.2008	-.2074	-.2162	-.2359
	111	-.2484	-.2159	-.1887	-.1722	-.1542	-.1658	-.1836	-.2107
	112	-.1756	-.1536	-.1436	-.1335	-.1144	-.1316	-.1379	-.1598
	113	-.1798	-.1700	-.1548	-.1390	-.1193	-.1113	-.0918	-.0821
	114	-.2669	-.2367	-.2106	-.1765	-.1580	-.1482	-.1474	-.1409
	115	-.2696	-.2329	-.2027	-.1825	-.1734	-.1887	-.1926	-.2181
	116	-.2192	-.1968	-.1854	-.1914	-.1870	-.2184	-.2480	-.2759
	117	-.3153	-.2726	-.2148	-.1752	-.1166	-.0610	-.0299	-.0087
	118	-.2847	-.2337	-.1994	-.1588	-.1272	-.1100	-.1039	-.0957
	119	-.2446	-.2116	-.1846	-.1603	-.1413	-.1346	-.1414	-.1496
	120	-.1926	-.1666	-.1491	-.1398	-.1386	-.1420	-.1606	-.1863
	121	-.1360	-.1297	-.1326	-.1344	-.1431	-.1753	-.2135	-.2472
	122	-.1592	-.1592	-.1126	-.0606	-.0109	.0298	.0713	.0942
	123	-.1589	-.1251	-.0831	-.0459	-.0039	.0243	.0513	.0595
	124	-.1322	-.0951	-.0711	-.0355	-.0176	.0078	.0301	.0478
	125	-.0907	-.0662	-.0477	-.0316	-.0160	.0051	.0178	.0420
	126	-.0667	-.0516	-.0369	-.0259	-.0194	-.0165	-.0080	-.0054
Aft fuselage area-rule addition	127	.0530	.0718	.0940	.1211	.1407	.1603	.1860	.2061
	128	.0311	.0534	.0742	.0903	.1140	.1249	.1013	.0618
	129	.0147	.0219	.0209	.0262	.0403	.0467	.0568	.0329
	130	-.0713	-.0576	-.0456	-.0256	-.0138	-.0228	-.0097	-.0168
	131	-.0574	-.0478	-.0440	-.0439	-.0308	-.0242	-.0270	-.0439
	132	-.0699	-.0554	-.0495	-.0454	-.0333	-.0239	-.0270	-.0413
	133	-.0746	-.0624	-.0552	-.0486	-.0382	-.0280	-.0187	-.0456
	134	-.0848	-.0779	-.0636	-.0607	-.0446	-.0409	-.0596	-.1179
	135	-.0895	-.0821	-.0700	-.0609	-.0505	-.0732	-.1214	-.1680
	136	-.0940	-.0793	-.0735	-.0596	-.0657	-.0762	-.0670	-.0669
	137	-.1240	-.1128	-.1055	-.1015	-.0938	-.0982	-.0903	-.0999
	138	-.1384	-.1262	-.1228	-.1196	-.1035	-.1015	-.1042	-.1185
	139	-.1203	-.1188	-.1097	-.1092	-.0975	-.0970	-.1026	-.1182
	140	-.1203	-.1105	-.1051	-.1001	-.0900	-.0905	-.1163	-.1592
	141	-.1138	-.1081	-.1020	-.0991	-.0916	-.1269	-.1549	-.1165
	142	-.1119	-.1042	-.0952	-.0913	-.0990	-.1092	-.0930	-.0996
	143	-.1099	-.1135	-.1176	-.1309	-.1278	-.1315	-.1349	-.1462
	144	-.0946	-.0924	-.0971	-.0987	-.0980	-.1025	-.1124	-.1306
	145	-.0749	-.0708	-.0755	-.0862	-.0810	-.0913	-.0944	-.1122
	146	-.0568	-.0613	-.0593	-.0642	-.0664	-.0747	-.0913	-.0818
	147	-.0766	-.0692	-.0711	-.0684	-.0708	-.0947	-.0665	-.0599

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(q) M = 0.90; β = 5.8°.

	Orifice number	C_p at α, deg, of —							
		-7.57	-5.05	-2.41	0.27	2.97	4.34	5.64	8.11
Fore fuselage area-rule addition	101	-.0067	-.0618	-.1148	-.1792	-.2330	-.2780	-.3212	-.4056
	102	-.0059	-.0382	-.0577	-.0722	-.0968	-.1117	-.1173	-.1494
	103	-.5177	-.4507	-.3757	-.3104	-.2678	-.2536	-.2181	-.1256
	104	-.4298	-.3166	-.2447	-.1700	-.1154	-.0867	-.0630	-.1517
	105	-.2292	-.2152	-.2079	-.2038	-.1962	-.1932	-.1870	-.2303
	106	-.2852	-.2490	-.2155	-.1951	-.1763	-.1661	-.1614	-.1379
	107	-.2167	-.1703	-.1344	-.0948	-.0561	-.0407	-.0363	-.0325
	108	-.1734	-.1843	-.1975	-.2054	-.2104	-.2111	-.2070	-.2098
	109	-.3369	-.2920	-.2540	-.2297	-.2219	-.2172	-.2155	-.2124
	110	-.3096	-.2673	-.2393	-.2164	-.2090	-.2169	-.2155	-.2219
	111	-.2543	-.2223	-.1989	-.1797	-.1771	-.1727	-.1714	-.1970
	112	-.1724	-.1625	-.1524	-.1380	-.1281	-.1273	-.1321	-.1414
	113	-.1797	-.1759	-.1583	-.1390	-.1209	-.1134	-.1008	-.0781
	114	-.2952	-.2482	-.2123	-.1829	-.1616	-.1482	-.1409	-.1354
	115	-.2648	-.2334	-.2106	-.1904	-.1766	-.1878	-.1862	-.1885
	116	-.2211	-.2057	-.1999	-.2043	-.2061	-.2094	-.2295	-.2529
	117	-.3619	-.2998	-.2519	-.1887	-.1325	-.1059	-.0789	-.0250
	118	-.3052	-.2611	-.2181	-.1755	-.1373	-.1276	-.1143	-.0962
	119	-.2714	-.2345	-.1992	-.1752	-.1514	-.1488	-.1407	-.1385
	120	-.1934	-.1842	-.1709	-.1594	-.1555	-.1480	-.1508	-.1620
	121	-.1358	-.1408	-.1450	-.1484	-.1590	-.1740	-.1906	-.2143
	122	-.1785	-.1480	-.1127	-.0651	-.0163	.0149	.0422	.0822
	123	-.1402	-.1211	-.0836	-.0428	-.0074	.0152	.0287	.0615
	124	-.1127	-.0915	-.0671	-.0349	-.0182	-.0031	.0126	.0385
	125	-.0764	-.0649	-.0454	-.0289	-.0157	-.0036	.0079	.0289
	126	-.0543	-.0475	-.0338	-.0255	-.0218	-.0146	-.0063	.0094
Aft fuselage area-rule addition	127	.0579	.0745	.1003	.1211	.1523	.1657	.1749	.1949
	128	.0346	.0578	.0771	.0981	.1235	.1373	.1348	.1179
	129	.0165	.0293	.0351	.0345	.0473	.0551	.0599	.0676
	130	-.0853	-.0505	-.0426	-.0237	-.0130	-.0165	-.0137	-.0052
	131	-.0724	-.0435	-.0434	-.0453	-.0341	-.0291	-.0247	-.0253
	132	-.0915	-.0549	-.0523	-.0462	-.0360	-.0288	-.0247	-.0273
	133	-.0950	-.0662	-.0549	-.0486	-.0393	-.0322	-.0220	-.0261
	134	-.1096	-.0706	-.0678	-.0561	-.0467	-.0388	-.0401	-.0674
	135	-.1255	-.0805	-.0704	-.0651	-.0577	-.0571	-.0687	-.1302
	136	-.1272	-.0893	-.0794	-.0627	-.0619	-.0673	-.0751	-.0739
	137	-.1462	-.1212	-.1124	-.1086	-.1001	-.0915	-.0896	-.1022
	138	-.1589	-.1352	-.1321	-.1266	-.1127	-.1066	-.1068	-.1179
	139	-.1530	-.1282	-.1228	-.1168	-.1075	-.0988	-.0965	-.1115
	140	-.1335	-.1224	-.1203	-.1094	-.1034	-.0944	-.0970	-.1315
	141	-.1337	-.1190	-.1165	-.1129	-.1053	-.1077	-.1244	-.1720
	142	-.1290	-.1159	-.1066	-.1030	-.1127	-.1161	-.1259	-.1074
	143	-.1801	-.1675	-.1725	-.1663	-.1633	-.1595	-.1596	-.1574
	144	-.1340	-.1290	-.1344	-.1322	-.1262	-.1313	-.1215	-.1178
	145	-.1109	-.1008	-.1058	-.1126	-.1075	-.1082	-.1033	-.1078
	146	-.0871	-.0925	-.0917	-.0901	-.0916	-.0879	-.0950	-.1072
	147	-.0861	-.0936	-.0962	-.0940	-.0994	-.1090	-.1204	-.0791

TABLE X – Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(r) $M = 0.95$; $\beta = 5.7^{\circ}$.

	Orifice number	C_p at α, deg, of —							
		-7.44	-4.95	-2.39	0.29	2.87	5.42	7.94	10.17
Fore fuselage area-rule addition	101	.0022	-.0568	-.1665	-.2532	-.2109	-.2895	-.3929	-.4169
	102	.0026	-.0440	-.0683	-.0299	-.0828	-.0975	-.1263	-.1419
	103	-.4430	-.4459	-.4190	-.2360	-.2598	-.2389	-.0964	-.1984
	104	-.4333	-.3898	-.1874	-.0950	-.1197	-.0776	-.1068	-.2635
	105	-.3088	-.2136	-.2526	-.2258	-.2150	-.1955	-.2081	-.1914
	106	-.1426	-.2199	-.2198	-.2050	-.1875	-.1606	-.1334	-.1266
	107	-.1244	-.1388	-.1317	-.1078	-.0587	-.0257	-.0192	-.0134
	108	-.1511	-.1761	-.1898	-.1797	-.1727	-.2163	-.2123	-.2067
	109	-.2858	-.2534	-.2476	-.1946	-.2349	-.2575	-.2531	-.2744
	110	-.2693	-.2477	-.2341	-.1883	-.2204	-.2530	-.2597	-.2758
	111	-.2413	-.2124	-.1946	-.1649	-.1899	-.1992	-.2215	-.2439
	112	-.1687	-.1560	-.1486	-.1361	-.1373	-.1418	-.1430	-.1402
	113	-.1631	-.1715	-.1532	-.1351	-.1136	-.0919	-.0661	-.0470
	114	-.3083	-.2728	-.2355	-.1945	-.1417	-.1272	-.1203	-.1070
	115	-.2960	-.2607	-.2365	-.2101	-.1775	-.1762	-.1789	-.1936
	116	-.2466	-.2297	-.2316	-.2188	-.2134	-.2282	-.2826	-.3291
	117	-.3463	-.3277	-.3146	-.2631	-.1761	-.0875	-.0178	.0177
	118	-.3509	-.3264	-.2963	-.2425	-.1712	-.1226	-.0949	-.0613
	119	-.3257	-.3115	-.2847	-.2355	-.1872	-.1483	-.1317	-.1258
	120	-.2539	-.2484	-.2413	-.2137	-.1795	-.1495	-.1515	-.1621
	121	-.1663	-.1852	-.1990	-.1866	-.1804	-.1931	-.2142	-.2338
	122	-.1571	-.1420	-.0978	-.0589	-.0083	.0458	.0977	.1351
	123	-.1182	-.1147	-.0761	-.0398	.0034	.0419	.0742	.0866
	124	-.1005	-.0758	-.0547	-.0301	-.0105	.0213	.0543	.0734
	125	-.0591	-.0494	-.0356	-.0233	-.0083	.0206	.0499	.0745
	126	-.0388	-.0318	-.0245	-.0180	-.0102	.0077	.0273	.0455
Aft fuselage area-rule addition	127	.0002	.0443	.0970	.1361	.1516	.1600	.1751	.2091
	128	-.0279	.0221	.0817	.1103	.1284	.1251	.1042	.0864
	129	.0270	.0310	.0422	.0425	.0523	.0265	.0086	.0088
	130	-.0691	-.0492	-.0332	-.0213	-.0085	-.0248	-.0160	-.0054
	131	-.0482	-.0434	-.0453	-.0420	-.0291	-.0381	-.0460	-.0434
	132	-.0651	-.0489	-.0538	-.0470	-.0294	-.0391	-.0485	-.0412
	133	-.0697	-.0603	-.0604	-.0489	-.0303	-.0367	-.0497	-.0461
	134	-.0765	-.0678	-.0703	-.0604	-.0431	-.0601	-.0916	-.1404
	135	-.0841	-.0757	-.0811	-.0654	-.0494	-.0776	-.1399	-.1395
	136	-.0954	-.0847	-.0797	-.0705	-.0623	-.0783	-.0801	-.0804
	137	-.1452	-.1203	-.1138	-.1129	-.0942	-.0814	-.0826	-.1182
	138	-.1637	-.1319	-.1279	-.1281	-.1117	-.0975	-.1155	-.1460
	139	-.1554	-.1265	-.1249	-.1199	-.1049	-.0873	-.1049	-.1474
	140	-.1452	-.1227	-.1192	-.1168	-.1039	-.1001	-.1479	-.2069
	141	-.1395	-.1219	-.1192	-.1163	-.1073	-.1226	-.1518	-.1452
	142	-.1341	-.1173	-.1134	-.1155	-.1098	-.1057	-.1164	-.1436
	143	-.2430	-.2313	-.2276	-.2237	-.2030	-.1742	-.1654	-.1801
	144	-.2111	-.1850	-.1910	-.1790	-.1643	-.1415	-.1351	-.1464
	145	-.1583	-.1492	-.1550	-.1506	-.1373	-.1188	-.1228	-.1498
	146	-.1264	-.1135	-.1278	-.1255	-.1136	-.1060	-.1210	-.0872
	147	-.1148	-.1183	-.1317	-.1221	-.1192	-.1221	-.0763	-.0795

[redacted]

TABLE X - Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(s) $M = 0.98$; $\beta = 5.7^{o}$.

	Orifice number	C_p at α, deg, of —						
		-4.91	-2.44	0.29	2.87	5.40	7.81	10.25
Fore fuselage area-rule addition	101	-.0465	-.1378	-.2228	-.2163	-.2731	-.3853	-.4088
	102	-.0190	-.0493	-.0410	-.0537	-.0717	-.0965	-.1172
	103	-.3905	-.3966	-.2608	-.2427	-.2422	-.0948	-.1311
	104	-.3453	-.2602	-.1087	-.0965	-.0751	-.0965	-.2707
	105	-.3030	-.2169	-.2040	-.2063	-.1845	-.1922	-.1740
	106	-.3266	-.2002	-.1911	-.1822	-.1450	-.1175	-.1029
	107	-.1506	-.0914	-.0834	-.0431	-.0086	-.0029	.0063
	108	-.0913	-.1722	-.2608	-.2736	-.2823	-.3079	-.3109
	109	-.1823	-.2362	-.2755	-.2691	-.2678	-.2776	-.2901
	110	-.1690	-.2348	-.2561	-.2463	-.2524	-.2643	-.2656
	111	-.1520	-.2048	-.2144	-.2054	-.1961	-.2182	-.2340
	112	-.1204	-.1516	-.1527	-.1432	-.1339	-.1335	-.1290
	113	-.1407	-.1410	-.1303	-.1135	-.0834	-.0542	-.0271
	114	-.2568	-.2270	-.1873	-.1599	-.1301	-.1320	-.0957
	115	-.2527	-.2237	-.2042	-.2533	-.2870	-.2916	-.3568
	116	-.2203	-.2263	-.2694	-.3092	-.3402	-.3812	-.3933
	117	-.3342	-.2822	-.2247	-.1676	-.0786	-.0002	.0493
	118	-.3151	-.2575	-.2148	-.1682	-.0964	-.0645	-.0374
	119	-.2901	-.2450	-.2068	-.1773	-.1155	-.0895	-.0882
	120	-.2330	-.2039	-.1815	-.1754	-.1156	-.1059	-.1167
	121	-.1791	-.1658	-.1619	-.1693	-.1431	-.1284	-.1373
	122	-.1092	-.0744	-.0249	.0156	.0636	.1092	.1494
	123	-.0737	-.0483	-.0070	.0253	.0640	.0850	.1024
	124	-.0507	-.0276	.0018	.0260	.0429	.0597	.0834
	125	-.0186	-.0062	.0086	.0199	.0408	.0564	.0905
	126	-.0057	.0071	.0119	.0143	.0286	.0450	.0685
Aft fuselage area-rule addition	127	.0504	.0983	.1478	.1645	.1773	.1912	.2238
	128	.0252	.0806	.1256	.1377	.1389	.1136	.1028
	129	.0270	.0478	.0616	.0519	.0330	.0118	-.0103
	130	-.0691	-.0293	-.0048	-.0058	-.0152	-.0103	-.0004
	131	-.0578	-.0413	-.0277	-.0450	-.0648	-.0832	-.0711
	132	-.0735	-.0508	-.0329	-.0501	-.0677	-.0824	-.0563
	133	-.0788	-.0515	-.0352	-.0457	-.0636	-.0769	-.0640
	134	-.0972	-.0715	-.0497	-.0597	-.0814	-.1286	-.1509
	135	-.1066	-.0808	-.0555	-.0661	-.1017	-.1458	-.1369
	136	-.1160	-.0836	-.0628	-.0736	-.0974	-.0967	-.0878
	137	-.1122	-.1132	-.1133	-.0938	-.1363	-.1497	-.1641
	138	-.1274	-.1308	-.1291	-.1091	-.1530	-.1696	-.1900
	139	-.1224	-.1299	-.1249	-.1016	-.1431	-.1645	-.1845
	140	-.1100	-.1269	-.1215	-.1028	-.1531	-.2065	-.2396
	141	-.1073	-.1320	-.1232	-.1115	-.1755	-.1933	-.1301
	142	-.1022	-.1223	-.1223	-.1169	-.1569	-.1701	-.1808
	143	-.2445	-.2344	-.2468	-.2246	-.1755	-.1799	-.3121
	144	-.2206	-.1957	-.2166	-.1880	-.1364	-.1543	-.2780
	145	-.1705	-.1763	-.1960	-.1643	-.1247	-.1468	-.2672
	146	-.1351	-.1419	-.1644	-.1335	-.1110	-.1376	-.1702
	147	-.1340	-.1398	-.1496	-.1340	-.1206	-.1049	-.1982

TABLE X - Continued

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(t) M = 0.99; $\beta = 5.7^{\circ}$.

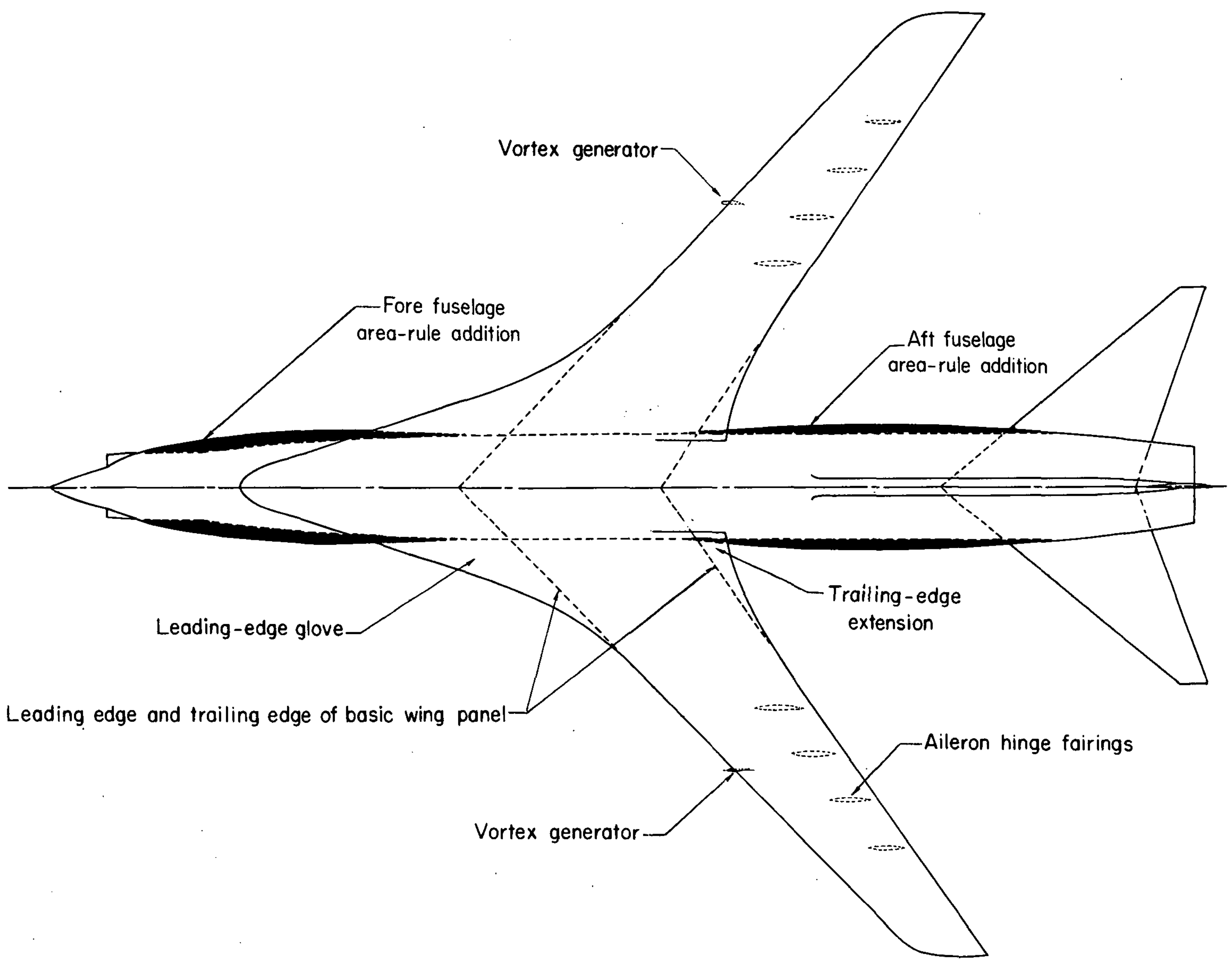
	Orifice number	C_p at α, deg, of —						
		-4.91	-2.39	0.23	2.85	5.38	7.85	10.24
Fore fuselage area-rule addition	101	-.0405	-.1170	-.1964	-.2997	-.2632	-.3679	-.4045
	102	-.0063	-.0426	-.0488	.0134	-.0599	-.0786	-.1308
	103	-.3788	-.3720	-.2495	-.1328	-.2227	-.1007	-.1818
	104	-.3242	-.2402	-.1171	.0052	-.0725	-.0847	-.2502
	105	-.2856	-.2235	-.1855	-.1859	-.1740	-.1771	-.1630
	106	-.3297	-.2102	-.1705	-.1701	-.1358	-.1035	-.0929
	107	-.1738	-.1080	-.0633	-.0400	.0028	.0088	.0171
	108	-.1970	-.2110	-.2422	-.2653	-.2777	-.2936	-.3129
	109	-.2902	-.2545	-.2565	-.2581	-.2588	-.2664	-.2775
	110	-.2612	-.2328	-.2367	-.2353	-.2453	-.2524	-.2536
	111	-.2245	-.1983	-.1962	-.1933	-.1869	-.2050	-.2231
	112	-.1610	-.1409	-.1339	-.1325	-.1234	-.1195	-.1157
	113	-.1233	-.1220	-.1276	-.1248	-.1056	-.0748	-.0384
	114	-.2140	-.2010	-.1976	-.1957	-.1946	-.1941	-.1586
	115	-.1992	-.2034	-.2382	-.2825	-.3162	-.3494	-.4004
	116	-.1857	-.2081	-.2937	-.3119	-.3417	-.3708	-.3861
	117	-.3069	-.2730	-.2183	-.1652	-.1115	.0124	.0650
	118	-.3004	-.2604	-.2171	-.1807	-.1435	-.0505	-.0060
	119	-.2791	-.2449	-.2085	-.1963	-.1842	-.0892	-.0414
	120	-.2198	-.2024	-.1928	-.1928	-.1941	-.1159	-.0714
	121	-.1663	-.1628	-.1698	-.2016	-.2433	-.2902	-.3253
	122	-.1182	-.0789	-.0311	.0317	.0767	.1268	.1597
	123	-.0854	-.0476	-.0085	.0426	.0784	.1035	.1079
	124	-.0583	-.0317	.0057	.0357	.0557	.0790	.0827
	125	-.0312	-.0113	.0167	.0360	.0563	.0685	.0929
	126	-.0136	.0001	.0238	.0321	.0434	.0605	.0805
Aft fuselage area-rule addition	127	.0661	.1038	.1482	.1652	.1801	.1980	.2320
	128	.0445	.0868	.1327	.1380	.1432	.1216	.1104
	129	.0330	.0599	.0711	.0554	.0425	.0300	.0037
	130	-.0557	-.0160	.0081	-.0002	-.0072	.0056	.0147
	131	-.0525	-.0275	-.0176	-.0416	-.0529	-.0724	-.0670
	132	-.0584	-.0402	-.0238	-.0410	-.0532	-.0687	-.0515
	133	-.0714	-.0472	-.0306	-.0446	-.0551	-.0699	-.0585
	134	-.0872	-.0632	-.0401	-.0530	-.0738	-.1173	-.1436
	135	-.0971	-.0699	-.0459	-.0632	-.0895	-.1331	-.1223
	136	-.1058	-.0726	-.0493	-.0679	-.0888	-.0825	-.0751
	137	-.1310	-.1078	-.1038	-.1185	-.1303	-.1380	-.1424
	138	-.1445	-.1254	-.1152	-.1318	-.1477	-.1604	-.1741
	139	-.1436	-.1191	-.1158	-.1291	-.1394	-.1564	-.1649
	140	-.1388	-.1186	-.1111	-.1253	-.1501	-.2038	-.2224
	141	-.1429	-.1223	-.1097	-.1330	-.1721	-.1866	-.1634
	142	-.1383	-.1214	-.1077	-.1352	-.1533	-.1651	-.1668
	143	-.2629	-.2650	-.2758	-.2647	-.2918	-.2968	-.3457
	144	-.2235	-.2383	-.2404	-.2327	-.2642	-.2793	-.3278
	145	-.1646	-.2037	-.2175	-.2149	-.2480	-.2716	-.3100
	146	-.1283	-.1751	-.1834	-.1789	-.2305	-.2565	-.2444
	147	-.1194	-.1662	-.1725	-.1721	-.2249	-.1956	-.2597

TABLE X - Concluded

PRESSURE DISTRIBUTIONS OVER SIDE FUSELAGE WITH AREA-RULE ADDITIONS ON

(u) M = 1.00; β = 5.7°.

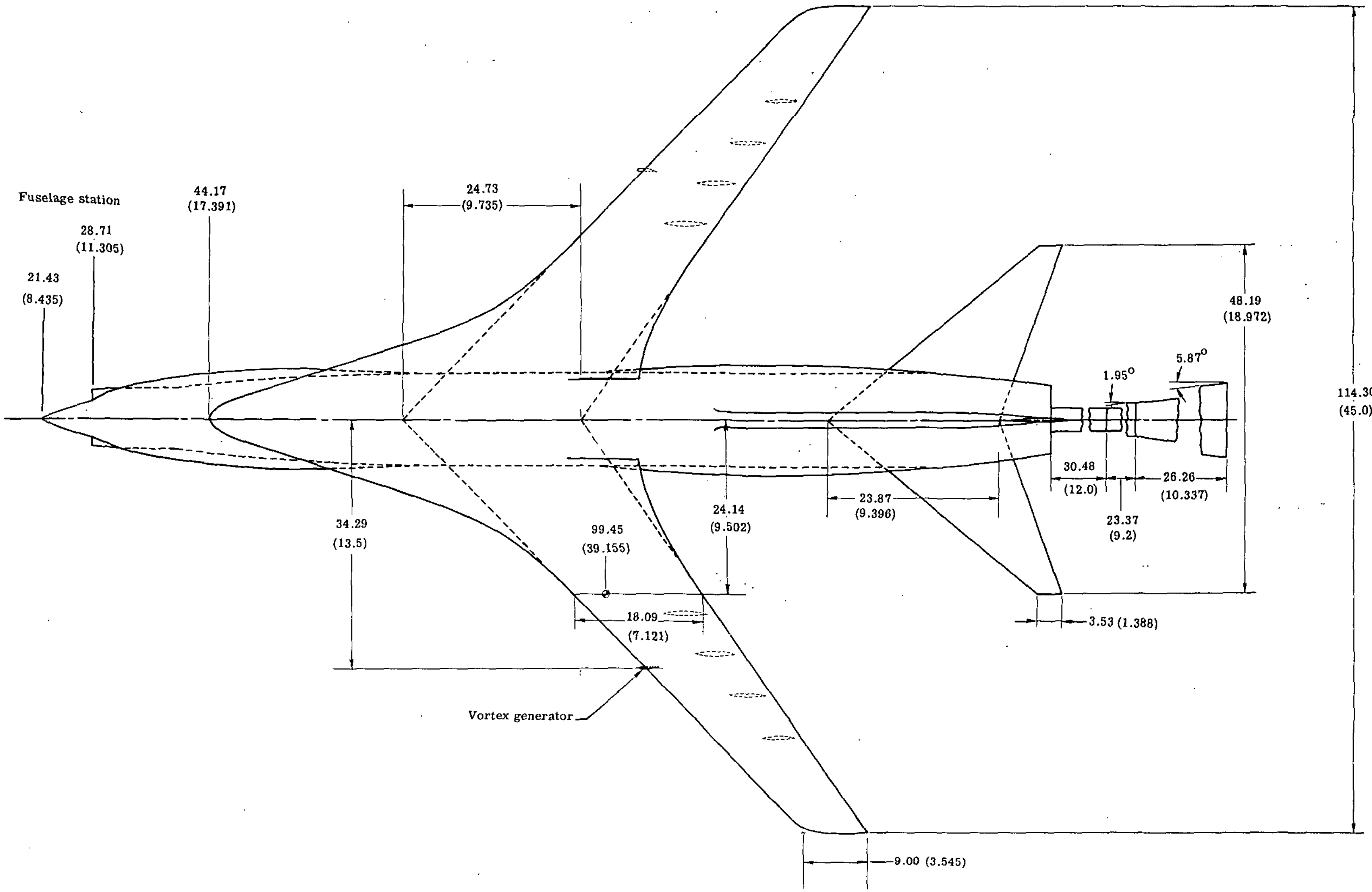
	Orifice number	C_p at α, deg, of —						
		-3.65	-2.38	0.22	2.83	5.34	7.84	10.22
Fore fuselage area-rule addition	101	-.0697	-.1094	-.1929	-.2853	-.2560	-.3604	-.3981
	102	-.0027	-.0246	-.0525	.0112	-.0566	-.0688	-.1185
	103	-.3567	-.3555	-.2666	-.1285	-.2036	-.0904	-.1754
	104	-.2814	-.2337	-.1204	.0064	-.0683	-.0833	-.2394
	105	-.2678	-.2283	-.1818	-.1730	-.1634	-.1649	-.1516
	106	-.2412	-.2112	-.1638	-.1580	-.1230	-.0915	-.0811
	107	-.1338	-.1032	-.0560	-.0353	.0122	.0224	.0294
	108	-.2114	-.2148	-.2301	-.2563	-.2683	-.2830	-.3082
	109	-.2695	-.2605	-.2453	-.2517	-.2484	-.2561	-.2723
	110	-.2500	-.2332	-.2272	-.2303	-.2325	-.2386	-.2471
	111	-.2079	-.1974	-.1858	-.1871	-.1767	-.1931	-.2168
	112	-.1469	-.1375	-.1249	-.1251	-.1123	-.1087	-.1085
	113	-.1256	-.1188	-.1278	-.1297	-.1179	-.0938	-.0690
	114	-.2119	-.2029	-.2011	-.2069	-.2112	-.2313	-.2384
	115	-.2120	-.2056	-.2414	-.2879	-.3150	-.3440	-.4085
	116	-.2177	-.2232	-.2898	-.3209	-.3313	-.3604	-.3828
	117	-.2739	-.2605	-.2142	-.1677	-.1191	-.0445	.0649
	118	-.2705	-.2498	-.2147	-.1779	-.1560	-.1323	-.0158
	119	-.2449	-.2339	-.2091	-.1958	-.1928	-.1742	-.0743
	120	-.1981	-.1957	-.1939	-.1917	-.2012	-.2231	-.1631
	121	-.1464	-.1582	-.1730	-.2006	-.2545	-.3438	-.4194
	122	-.1089	-.0877	-.0442	.0107	.0738	.1402	.1743
	123	-.0801	-.0606	-.0166	.0280	.0806	.1237	.1176
	124	-.0511	-.0369	-.0033	.0262	.0636	.0948	.0848
	125	-.0249	-.0174	.0085	.0241	.0564	.0887	.0972
	126	-.0126	-.0046	.0139	.0214	.0500	.0776	.0904
Aft fuselage area-rule addition	127	.0885	.1037	.1389	.1732	.1876	.2047	.2391
	128	.0740	.0836	.1190	.1423	.1515	.1297	.1154
	129	.0632	.0603	.0679	.0555	.0438	.0348	.0168
	130	-.0256	-.0159	.0113	.0027	.0005	.0154	.0268
	131	-.0317	-.0232	-.0074	-.0351	-.0445	-.0606	-.0603
	132	-.0414	-.0340	-.0101	-.0349	-.0486	-.0589	-.0523
	133	-.0484	-.0386	-.0145	-.0404	-.0491	-.0586	-.0564
	134	-.0649	-.0561	-.0283	-.0525	-.0673	-.1107	-.1366
	135	-.0772	-.0660	-.0314	-.0646	-.0865	-.1216	-.1068
	136	-.0844	-.0616	-.0350	-.0689	-.0799	-.0744	-.0681
	137	-.1173	-.0926	-.0829	-.1014	-.1187	-.1293	-.1315
	138	-.1263	-.1090	-.1040	-.1195	-.1373	-.1519	-.1587
	139	-.1290	-.1111	-.0988	-.1166	-.1279	-.1483	-.1522
	140	-.1244	-.1088	-.0984	-.1140	-.1375	-.1911	-.2072
	141	-.1343	-.1099	-.0957	-.1222	-.1656	-.1734	-.1515
	142	-.1317	-.1082	-.0989	-.1300	-.1398	-.1476	-.1536
	143	-.2723	-.2552	-.2541	-.2737	-.2836	-.2893	-.3370
	144	-.2471	-.2311	-.2323	-.2466	-.2560	-.2655	-.3235
	145	-.2113	-.2008	-.2105	-.2272	-.2439	-.2633	-.3046
	146	-.1743	-.1665	-.1813	-.2007	-.2289	-.2449	-.2404
	147	-.1796	-.1721	-.1653	-.1927	-.2255	-.1957	-.2534



(a) General planform arrangement of 0.087-scale model.

Figure 1.- Model details. Linear dimensions are in centimeters (inches).

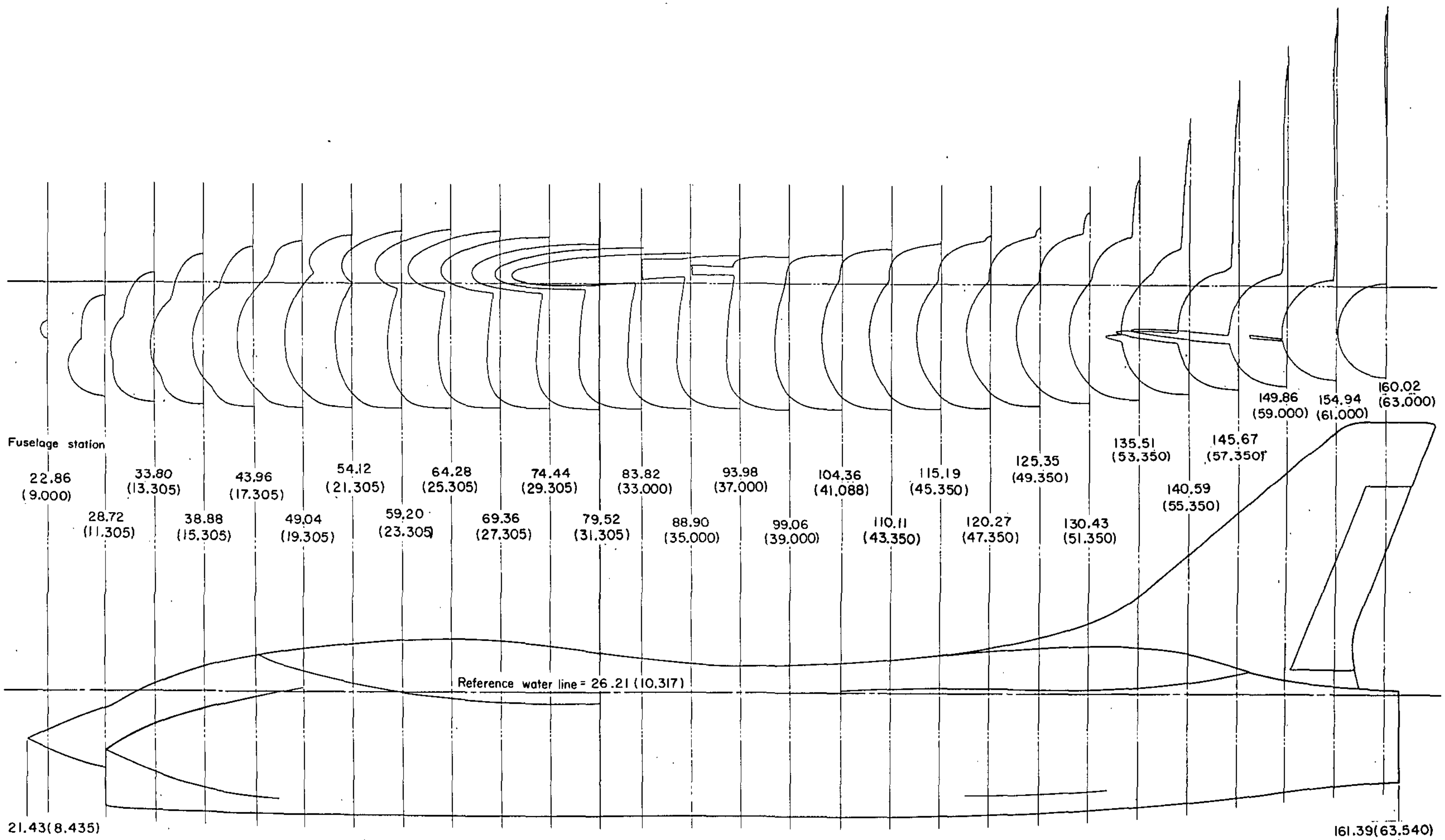
[redacted]



(a) Concluded.

Figure 1.- Continued.

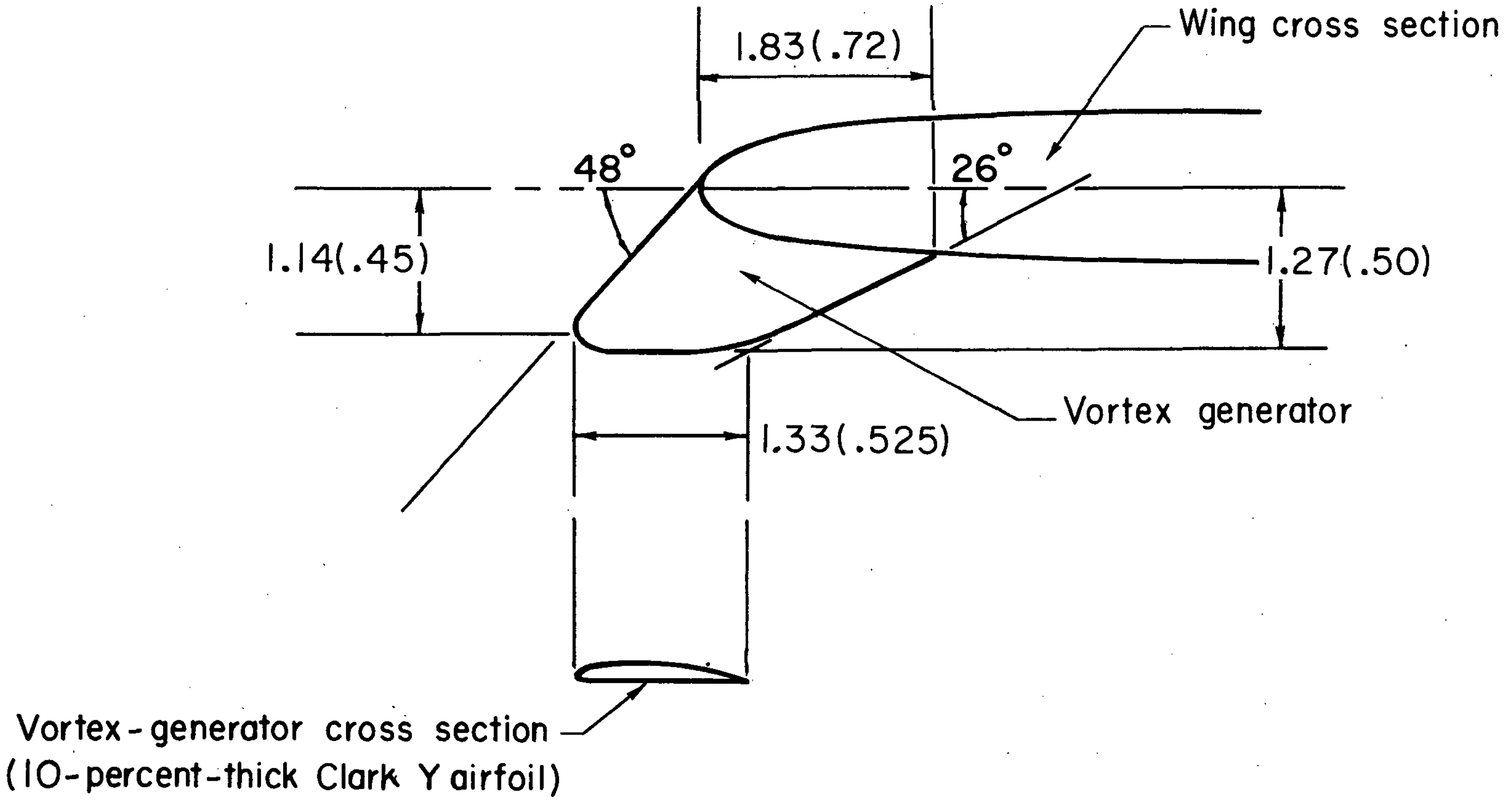
[redacted]



(b) Cross-section layout of 0.087-scale model with fuselage area-rule additions.

Figure 1.- Continued.

[redacted]

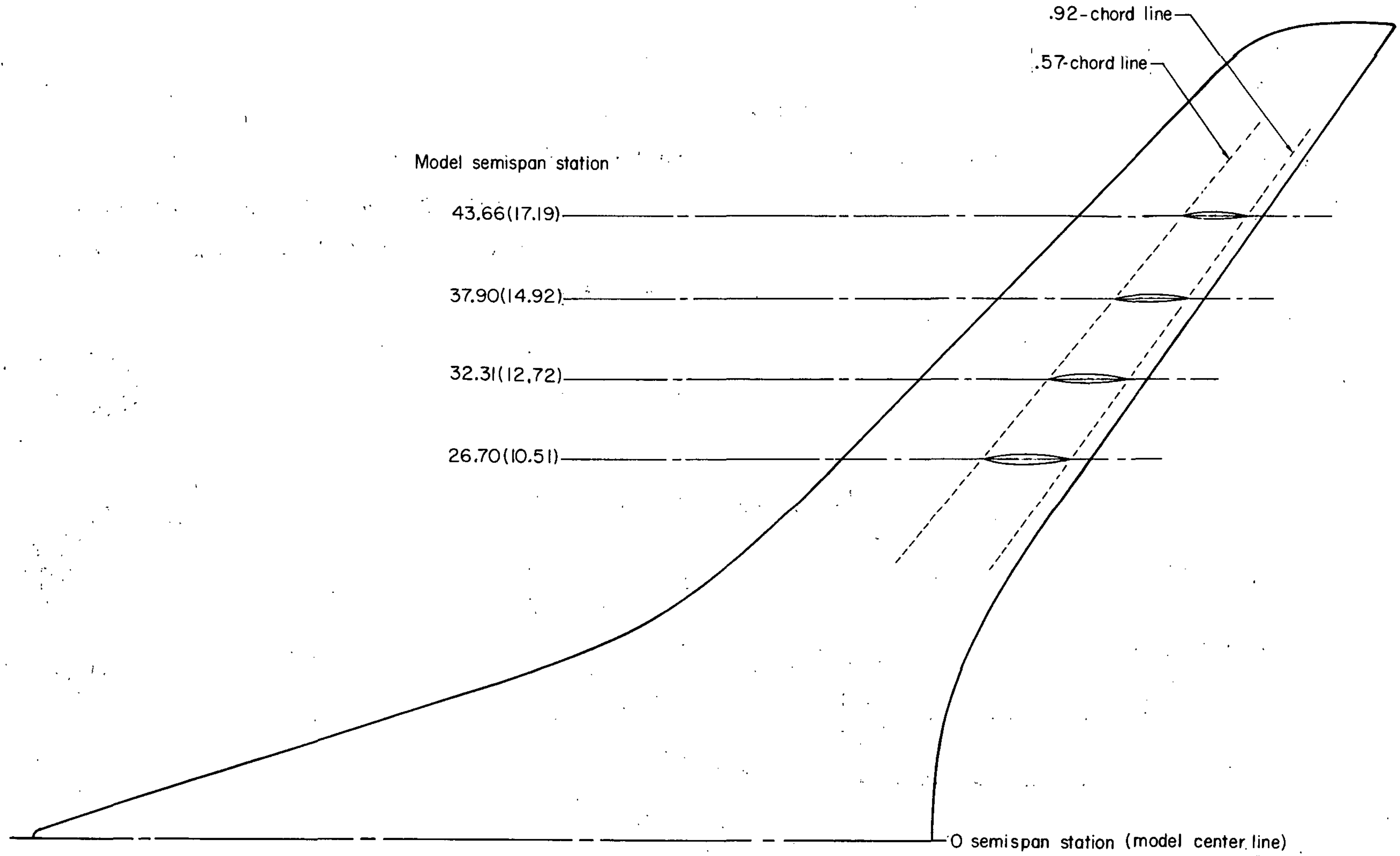


(c) Sketch of underwing leading-edge vortex generator.

Figure 1.- Continued.

[redacted]

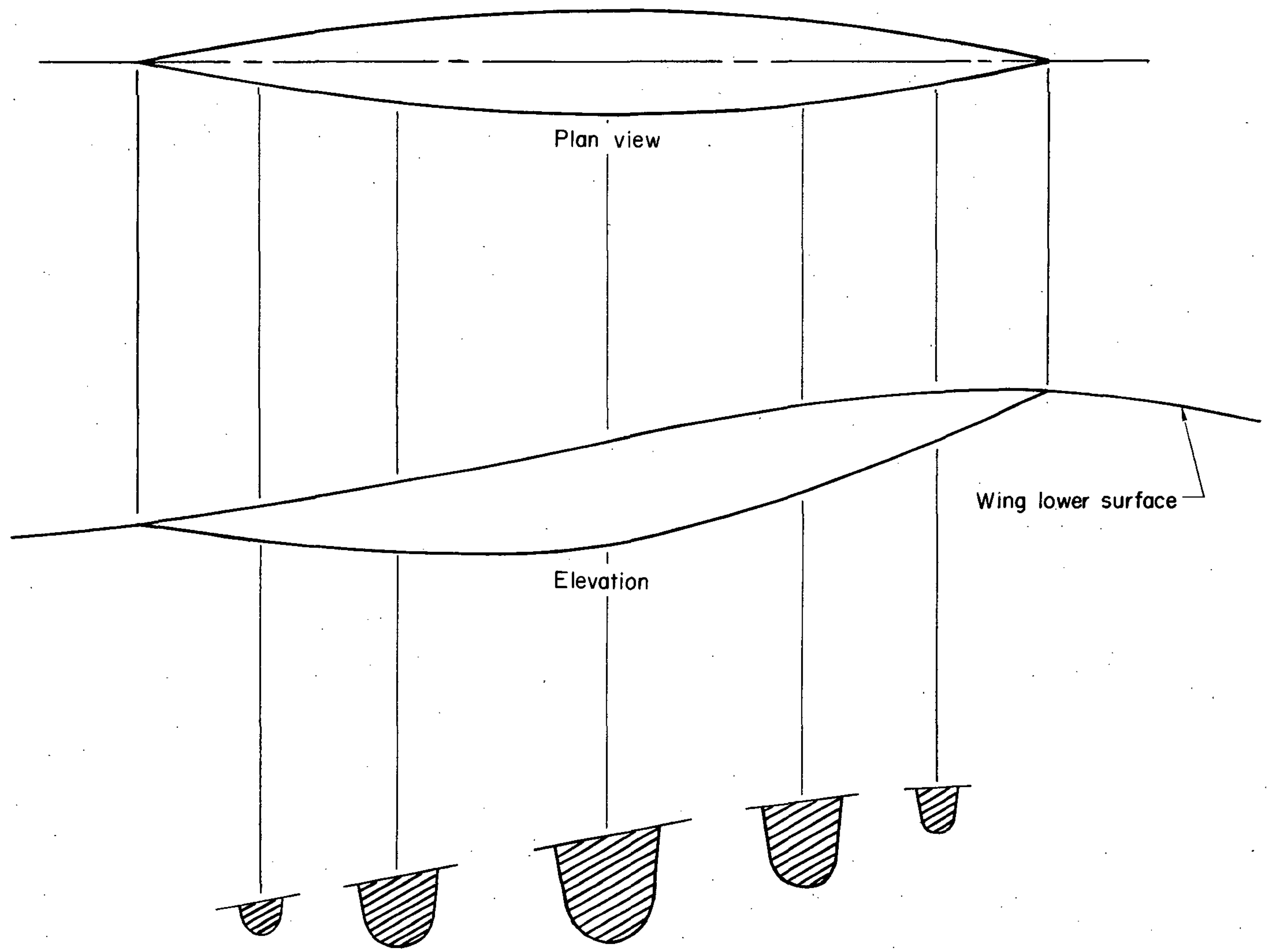
[redacted]



(d) Location of aileron hinge fairings.

Figure 1.- Continued.

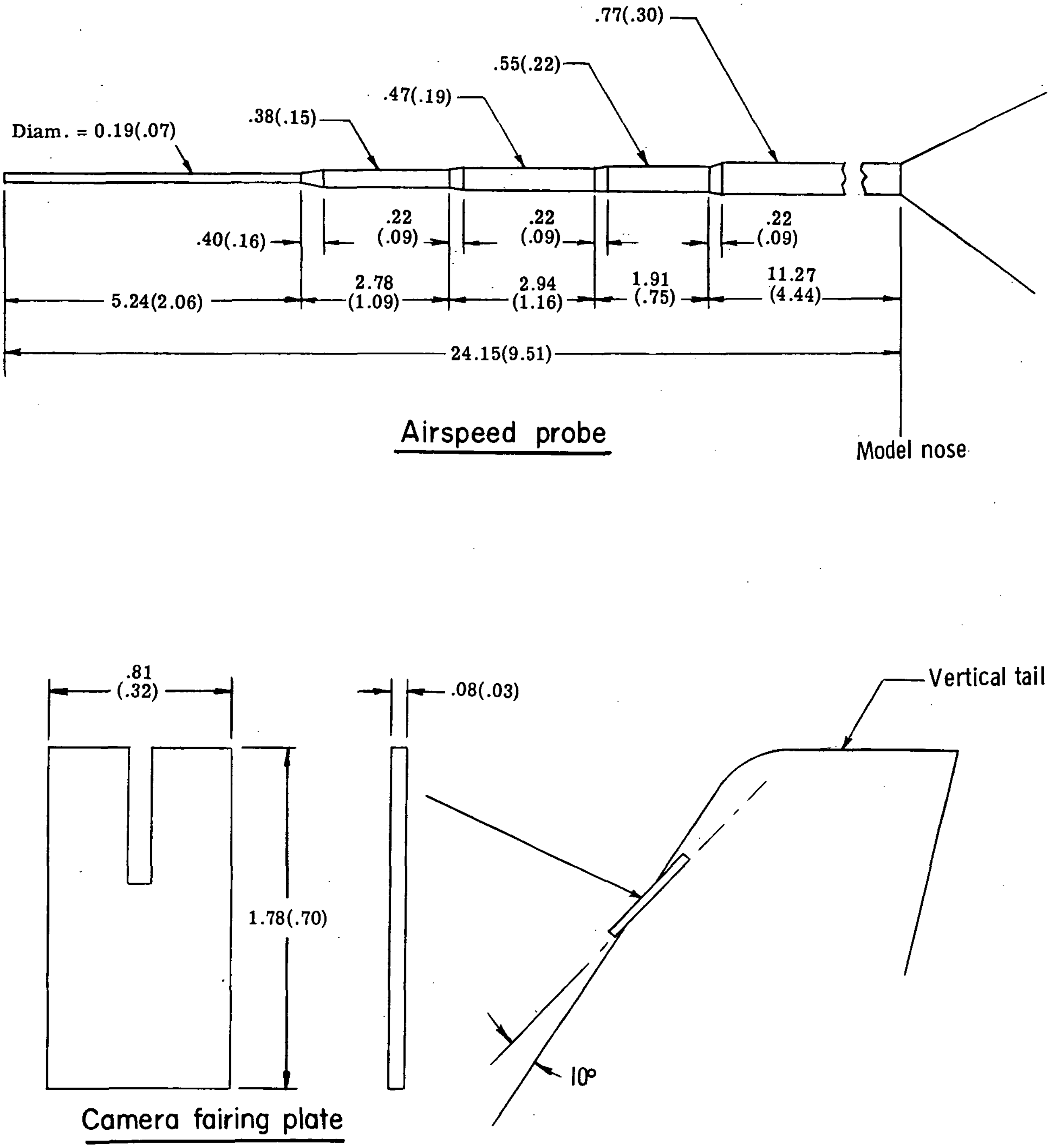
[redacted]



(e) Sketch of typical aileron hinge fairing.

Figure 1.- Continued.

[redacted]

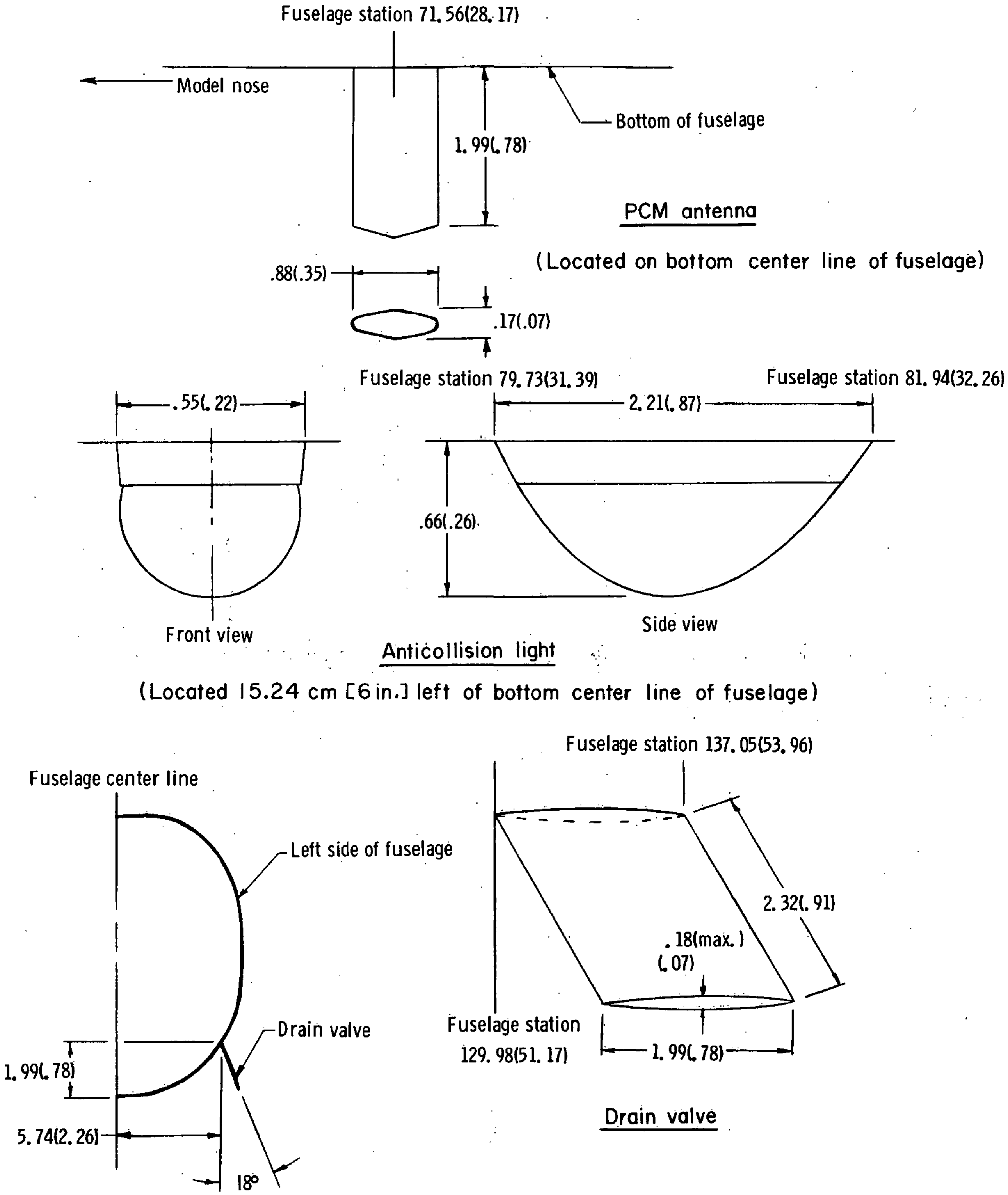


(f) Simulated full-scale airplane protuberances.

Figure 1.- Continued.

[redacted]

[redacted]



(f) Concluded.

Figure 1.- Continued.

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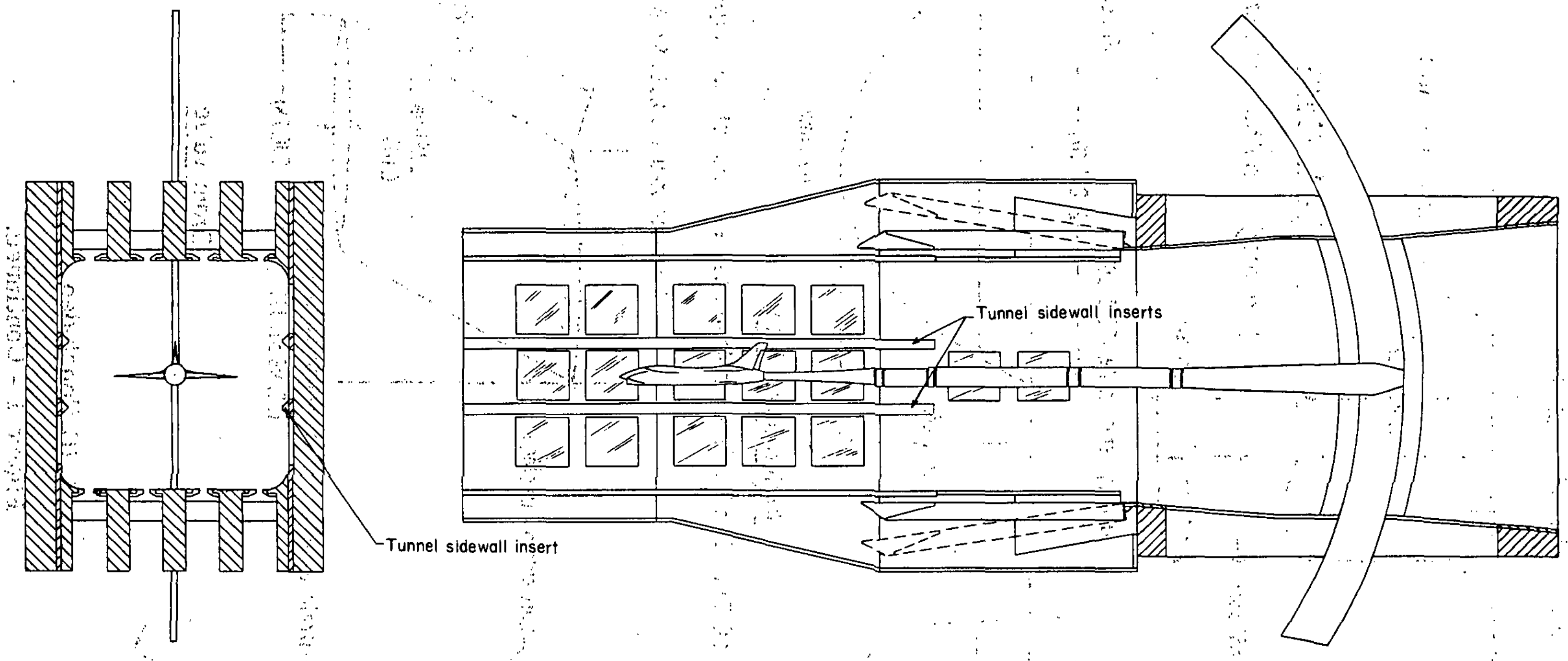
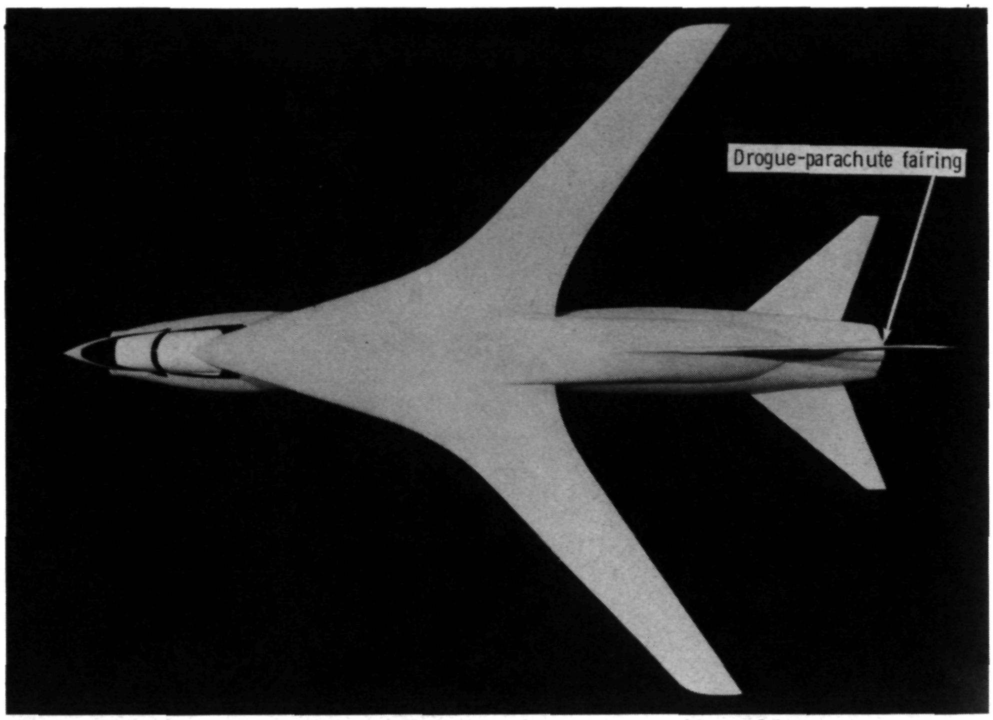


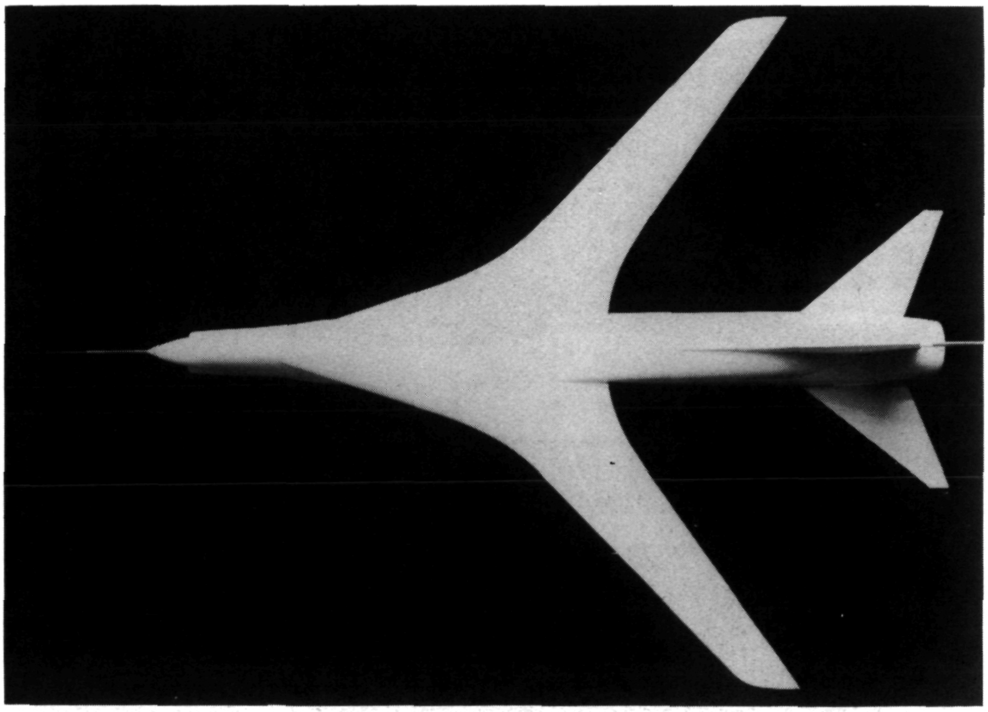
(g) Tunnel test section with sidewall inserts.

Figure 1.- Concluded.



Top view of model with fuselage area-rule additions

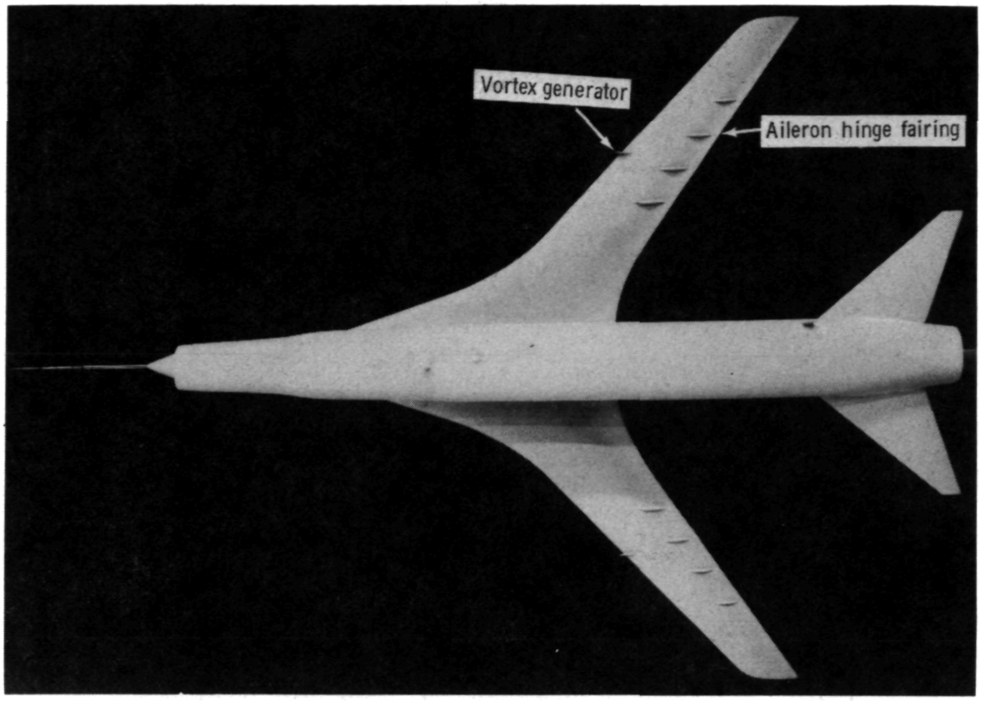
L-70-3988.1



Top view of model without fuselage area-rule additions

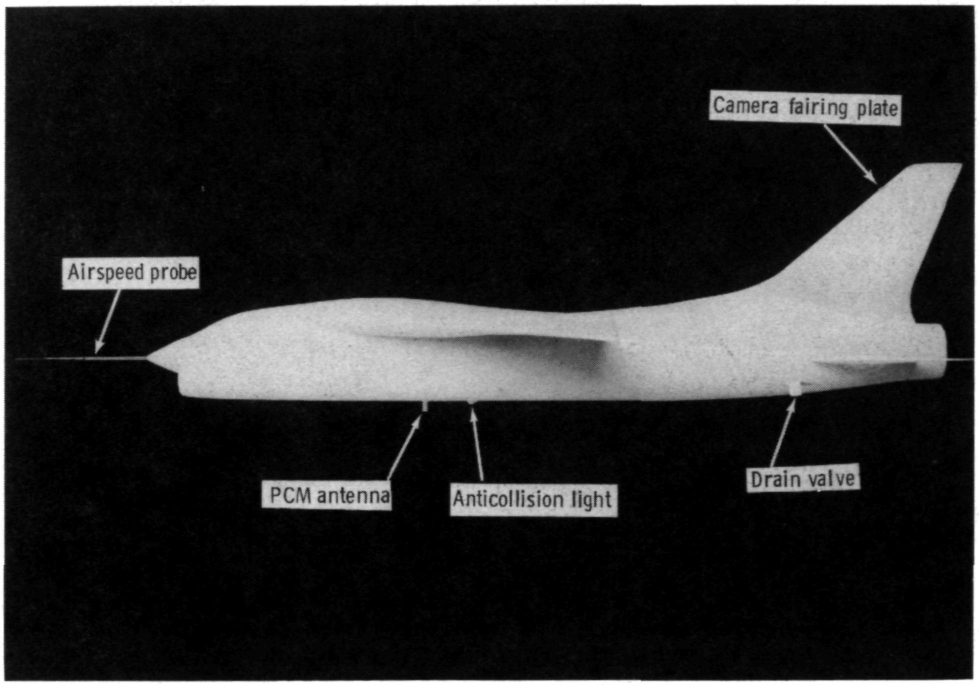
L-71-6987

Figure 2.- Photographs of 0.087-scale wind-tunnel model.



Bottom view of model without fuselage area-rule additions

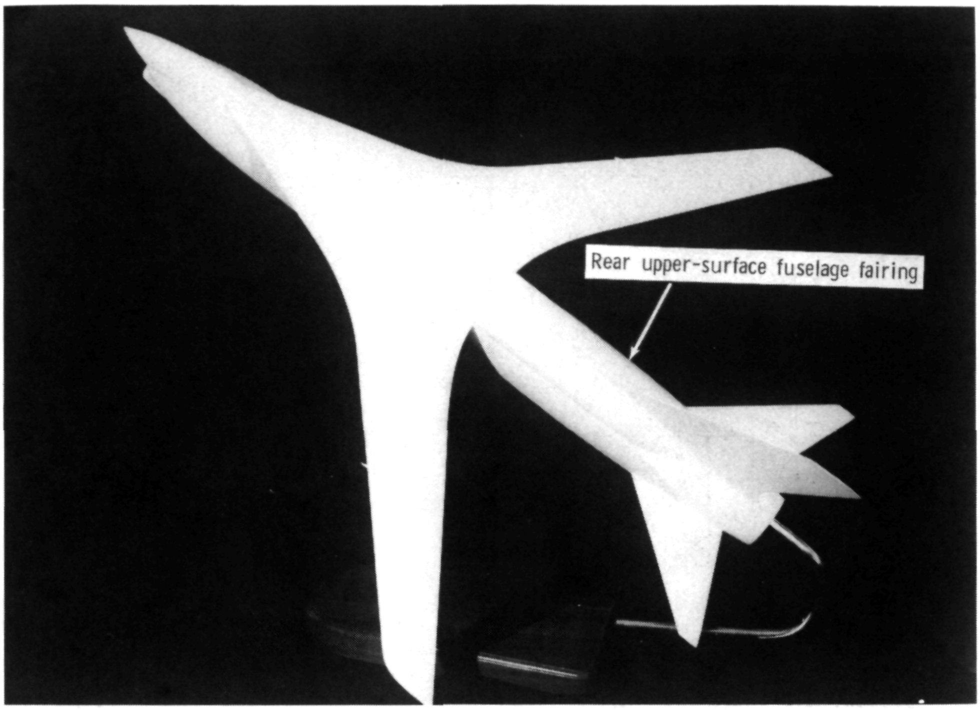
L-71-6992.1



Side view of model without fuselage area-rule additions

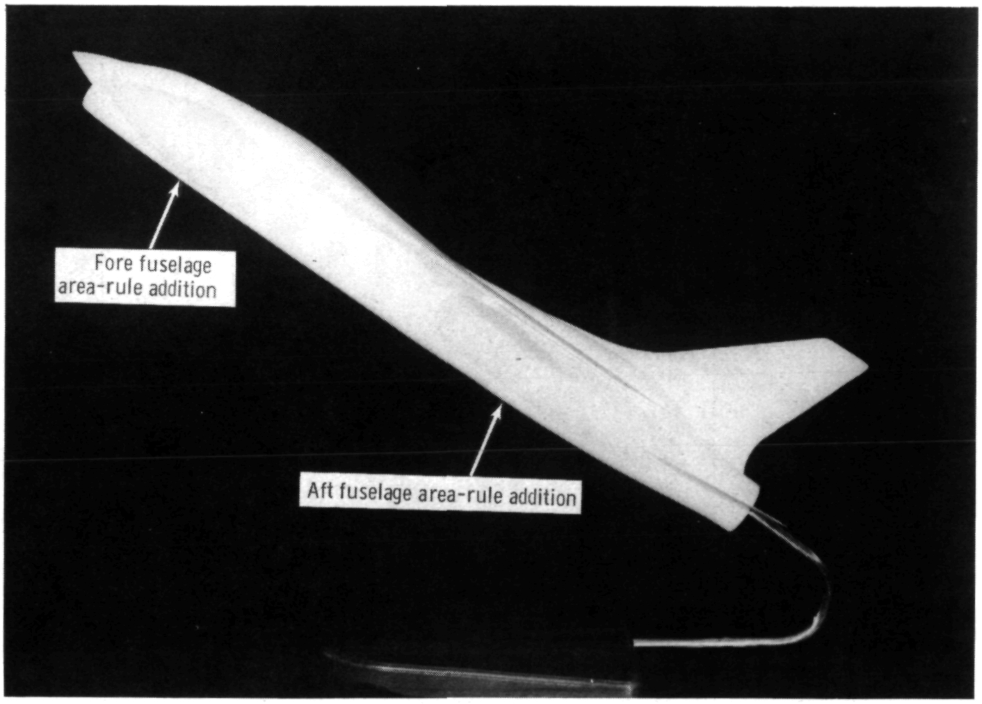
L-71-6991.1

Figure 2.- Continued.



Three-quarter top view of model with fuselage area-rule additions

L-72-1117.1



Side view of model with fuselage area-rule additions

L-72-1119.1

Figure 2.- Concluded.

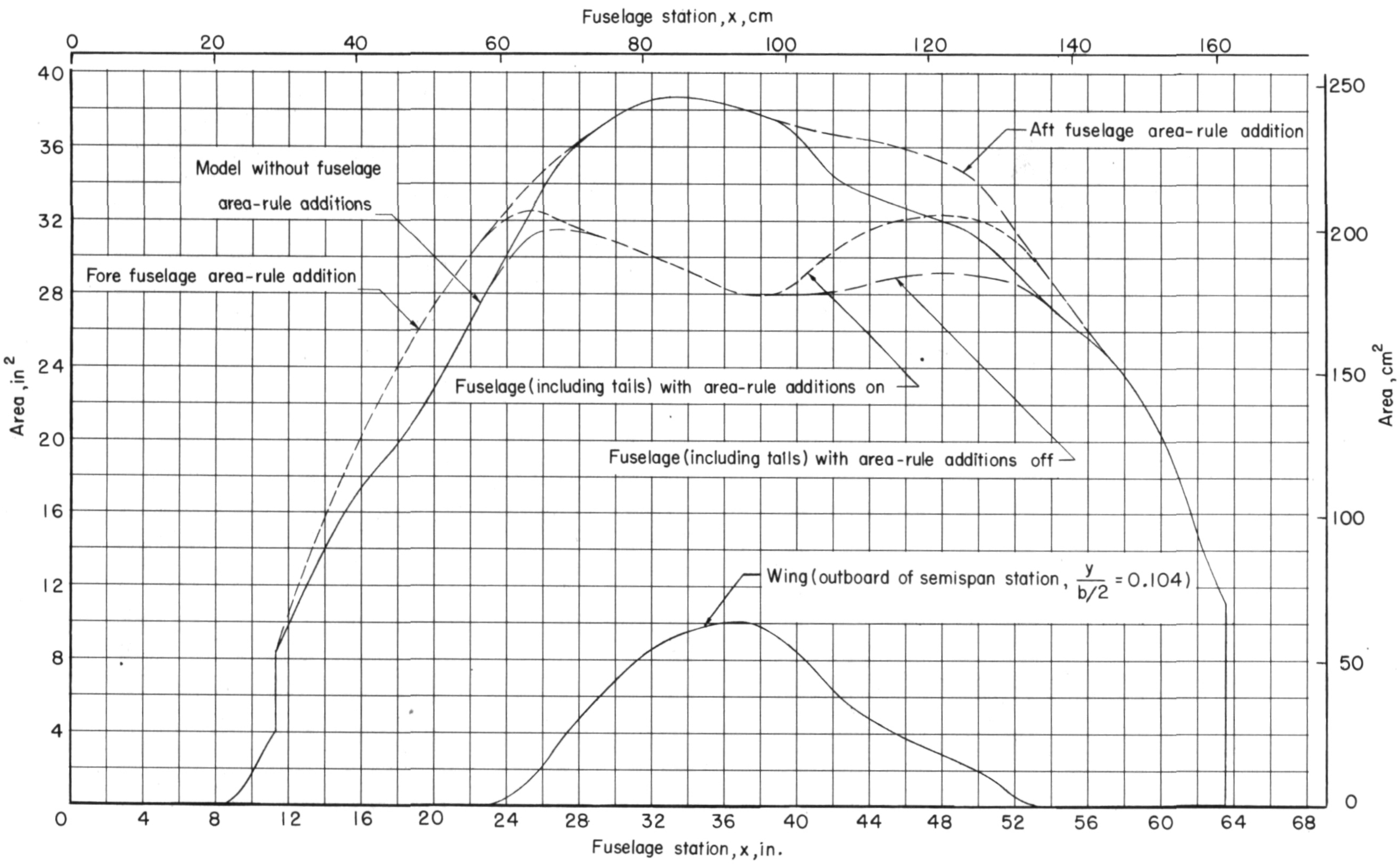


Figure 3.- Longitudinal progression of cross-sectional area taken normal to fuselage center line. (Model and fuselage areas include 28.39 cm^2 (4.40 in^2) of inlet area.)

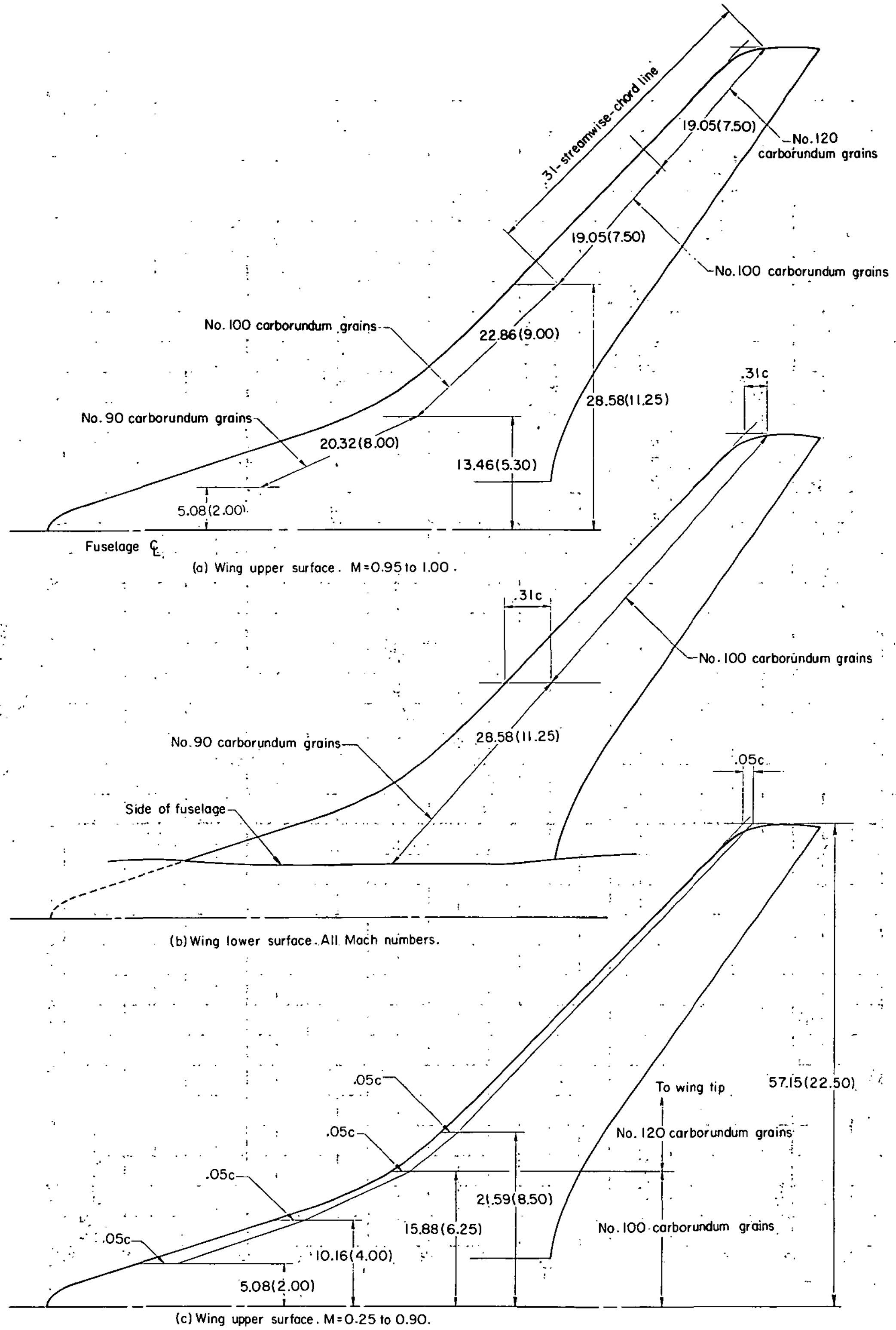
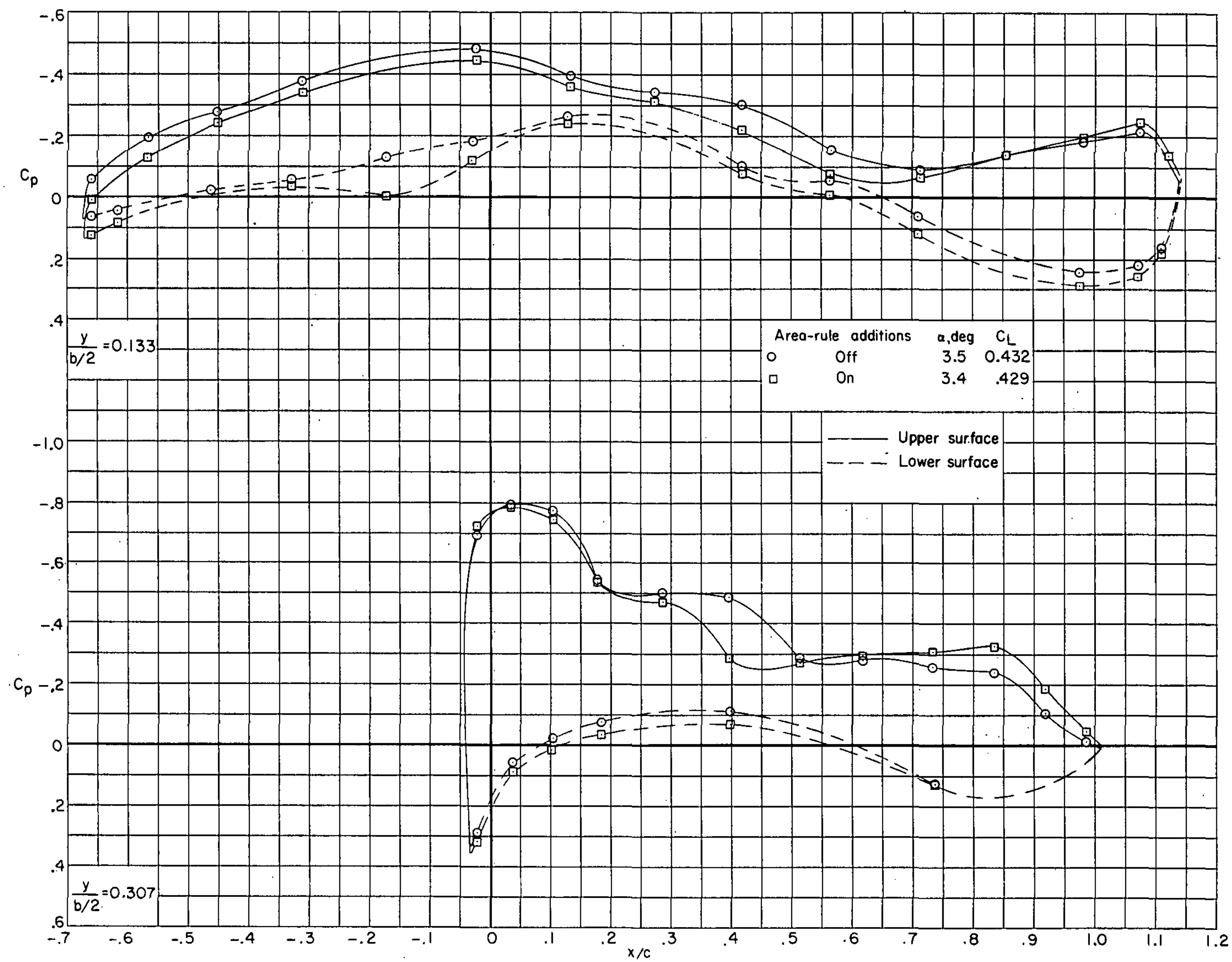


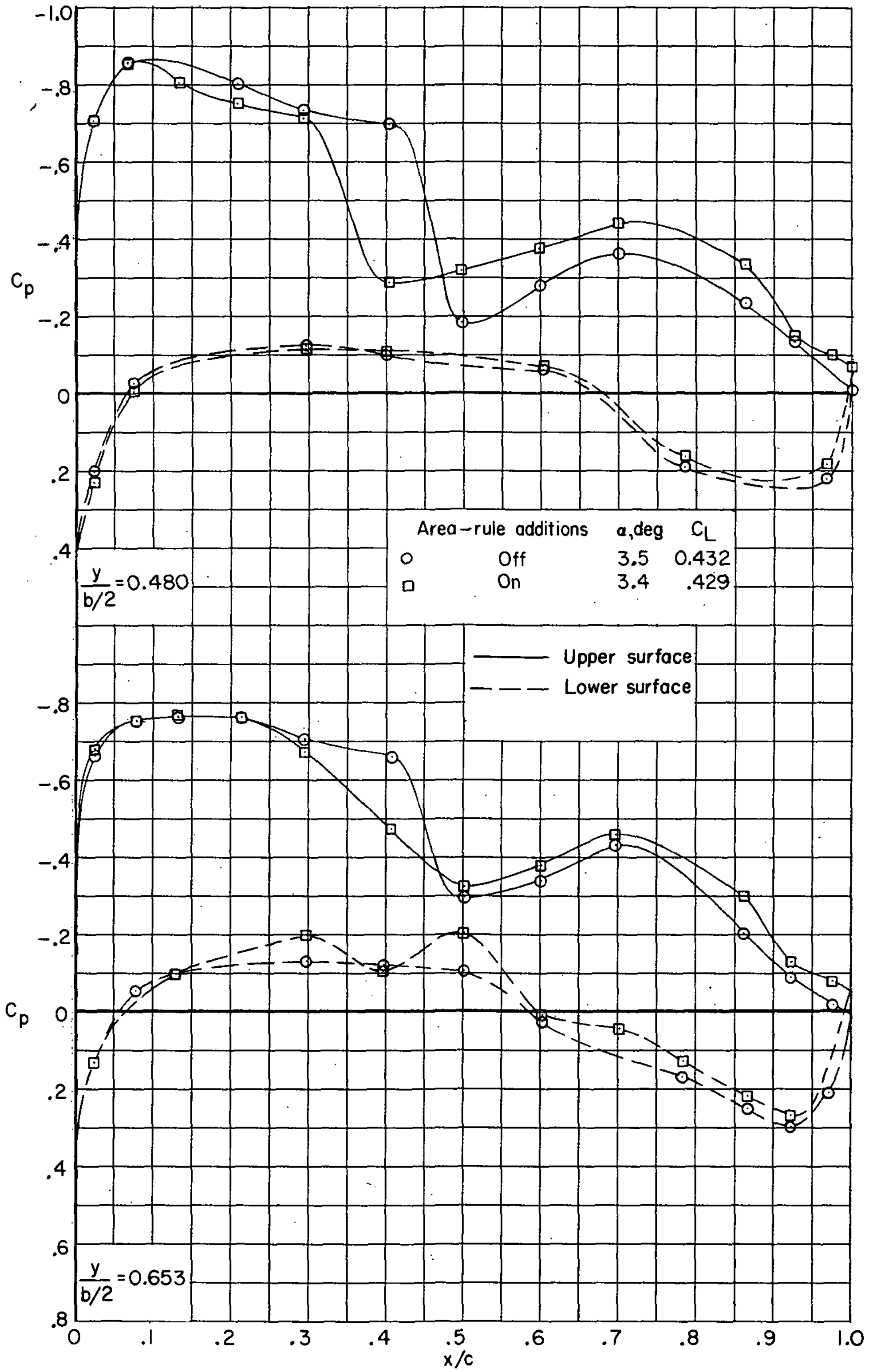
Figure 4.- Boundary-layer trip arrangements. Dimensions are in centimeters (inches).



(a) M = 0.97.

Figure 5.- Effect of fuselage area-rule additions on wing streamwise pressure distributions near cruise lift coefficient. $\beta = 0^{\circ}$; $\delta_h = -2.5^{\circ}$.

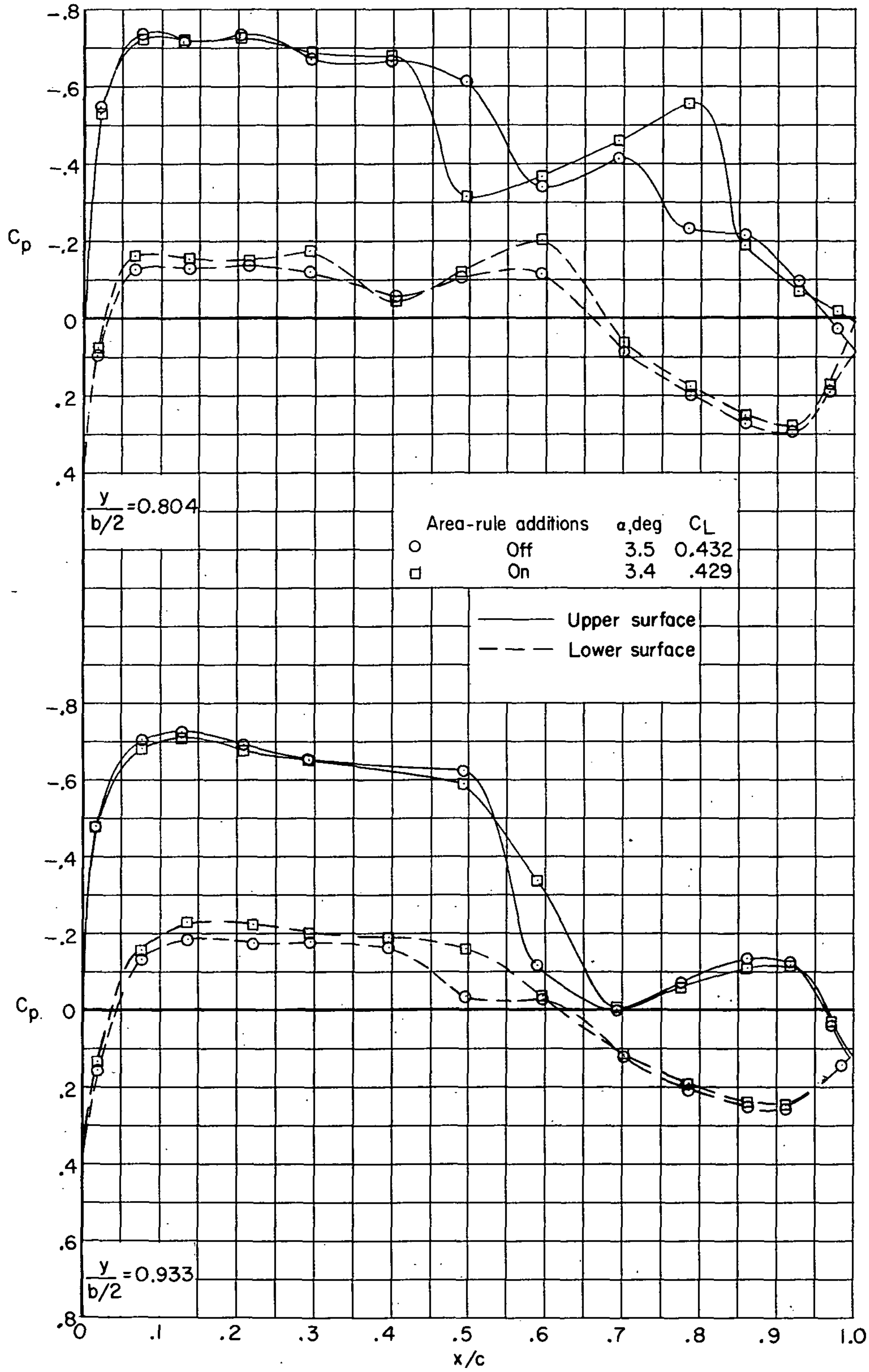
[redacted]



(a) M = 0.97. Continued.

Figure 5.- Continued.

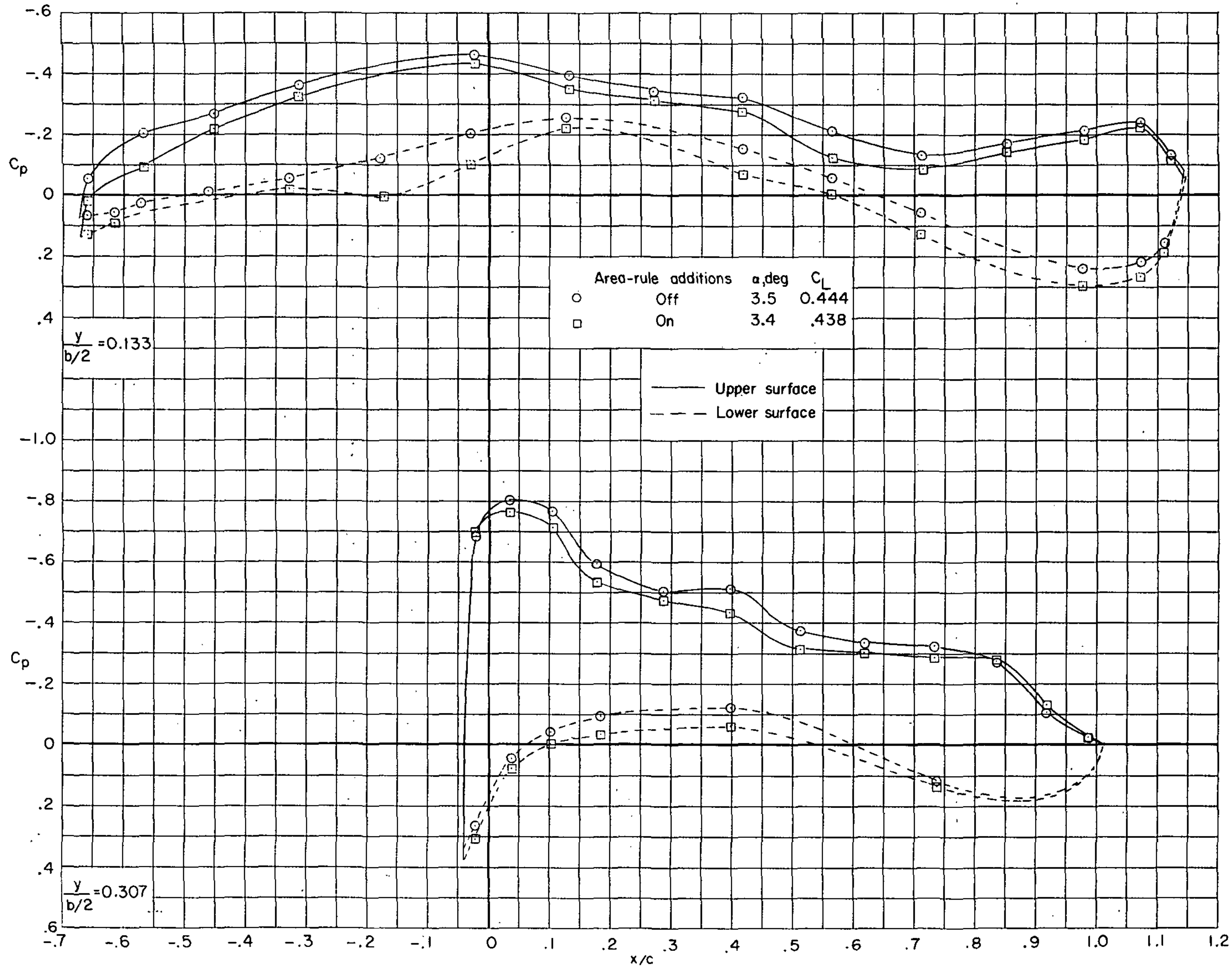
[redacted]



(a) M = 0.97. Concluded.

Figure 5.- Continued.

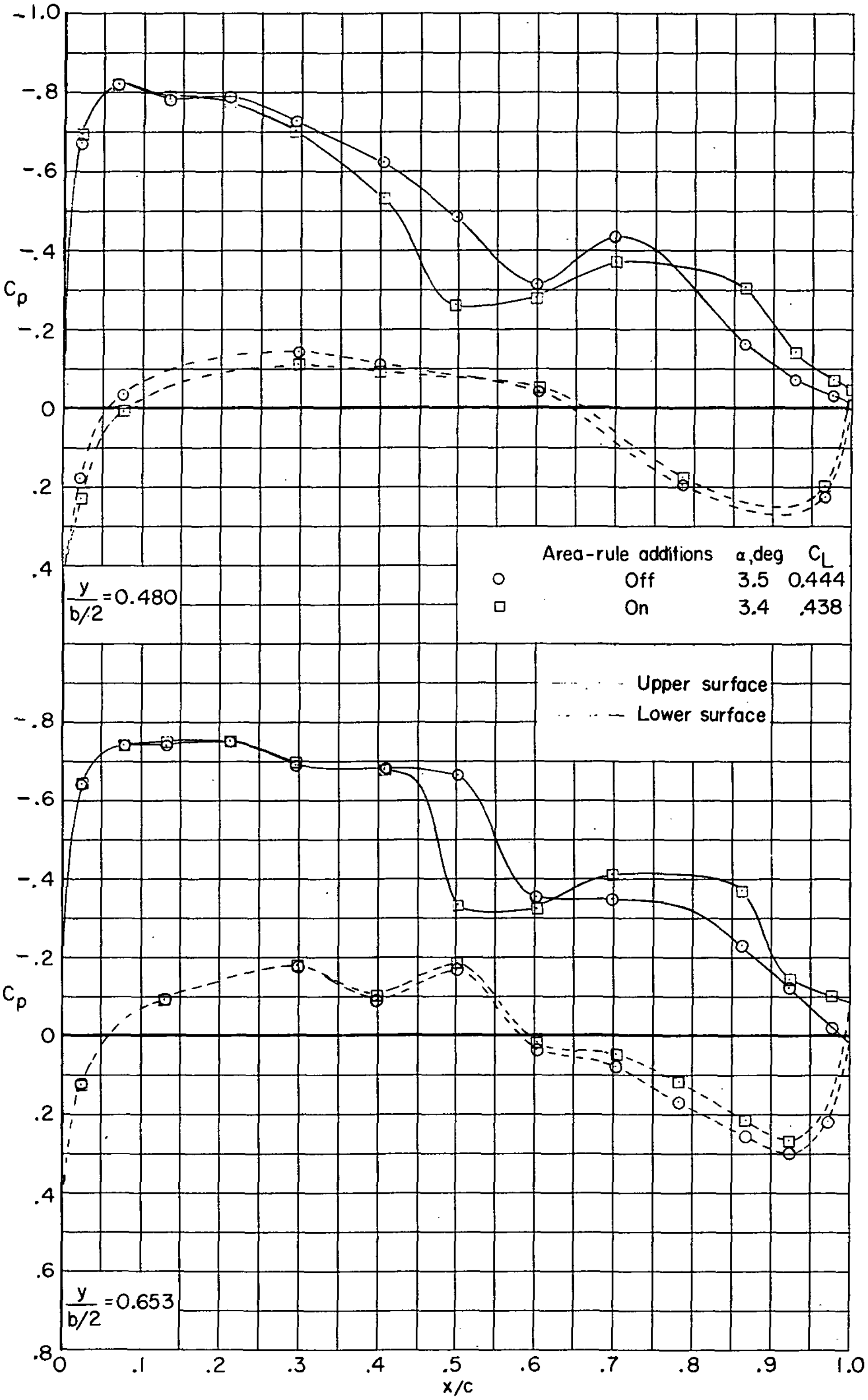
[redacted]



(b) M = 0.98.

Figure 5.- Continued.

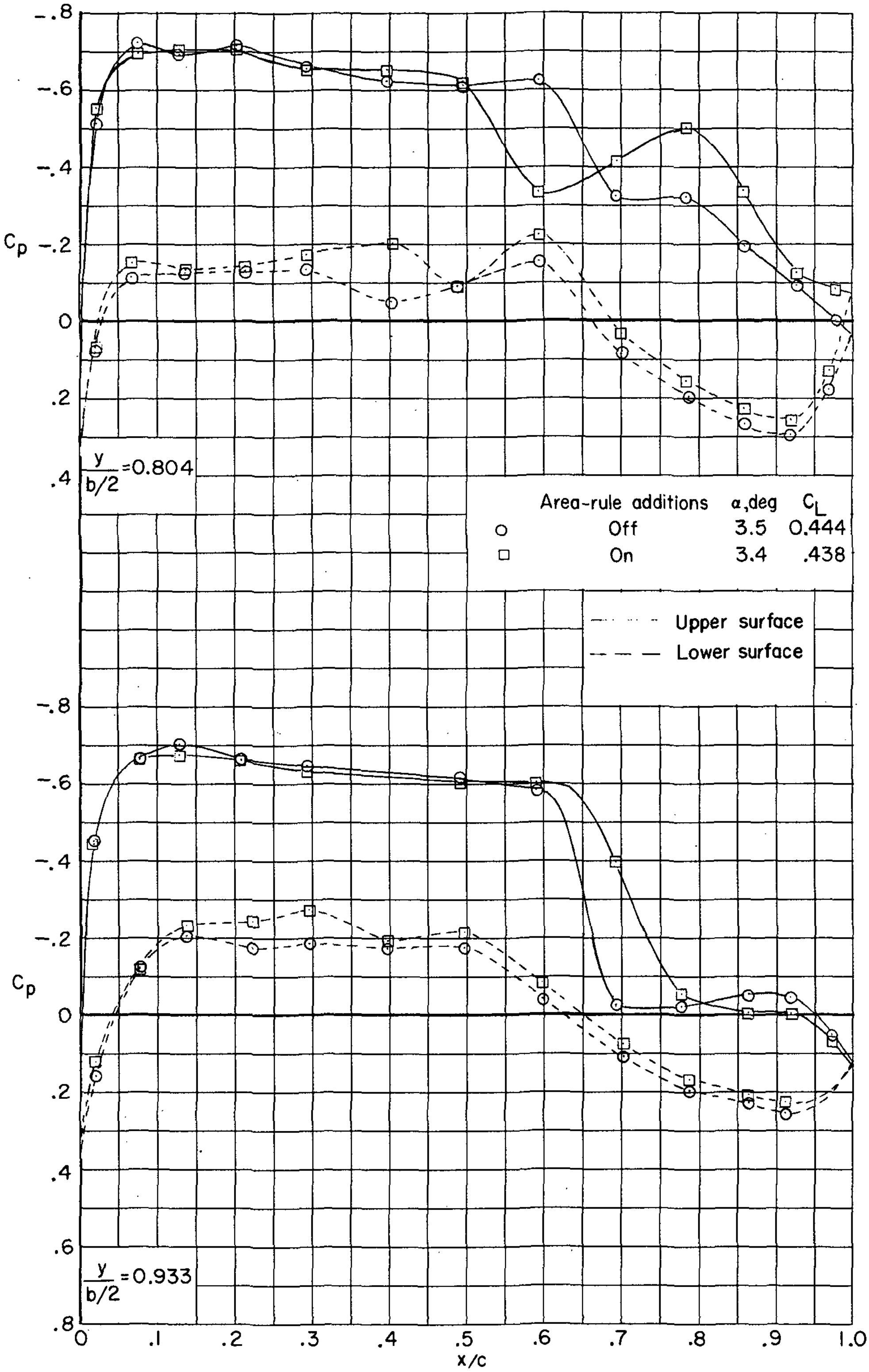
[redacted]



(b) M = 0.98. Continued.

Figure 5.- Continued.

[redacted]

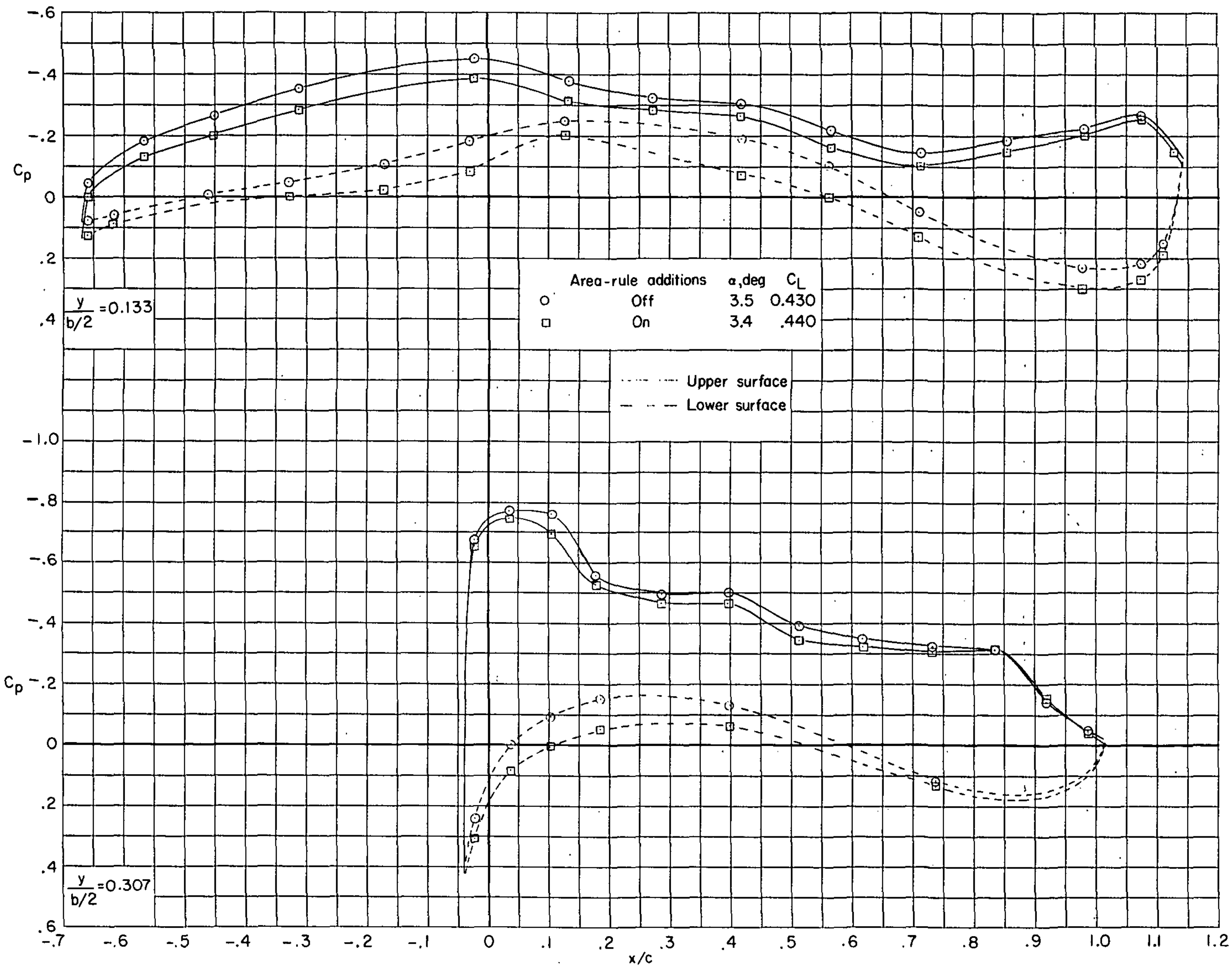


(b) M = 0.98. Concluded.

Figure 5.- Continued.

[redacted]

[redacted]

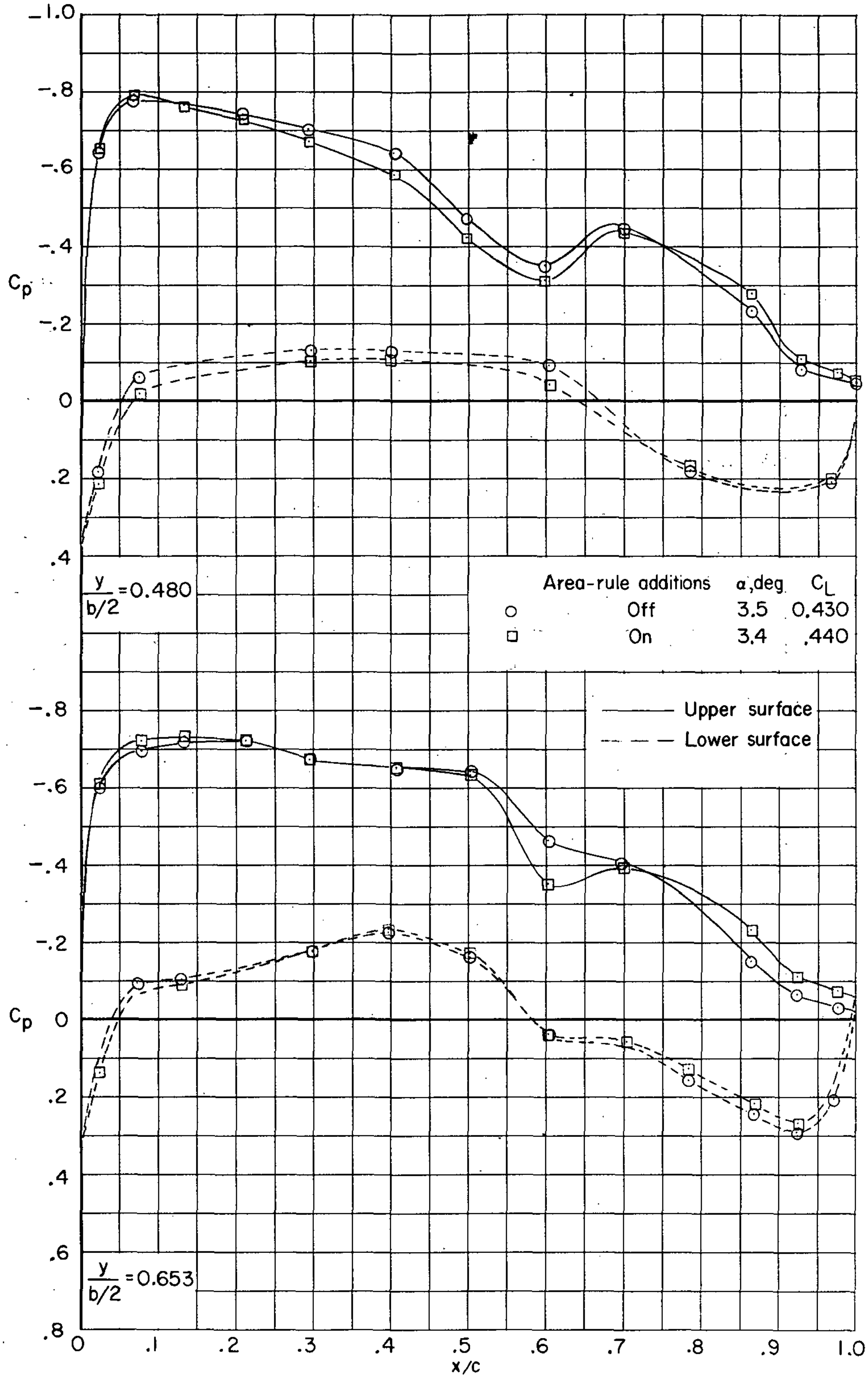


(c) M = 0.99.

Figure 5.- Continued.

[redacted]

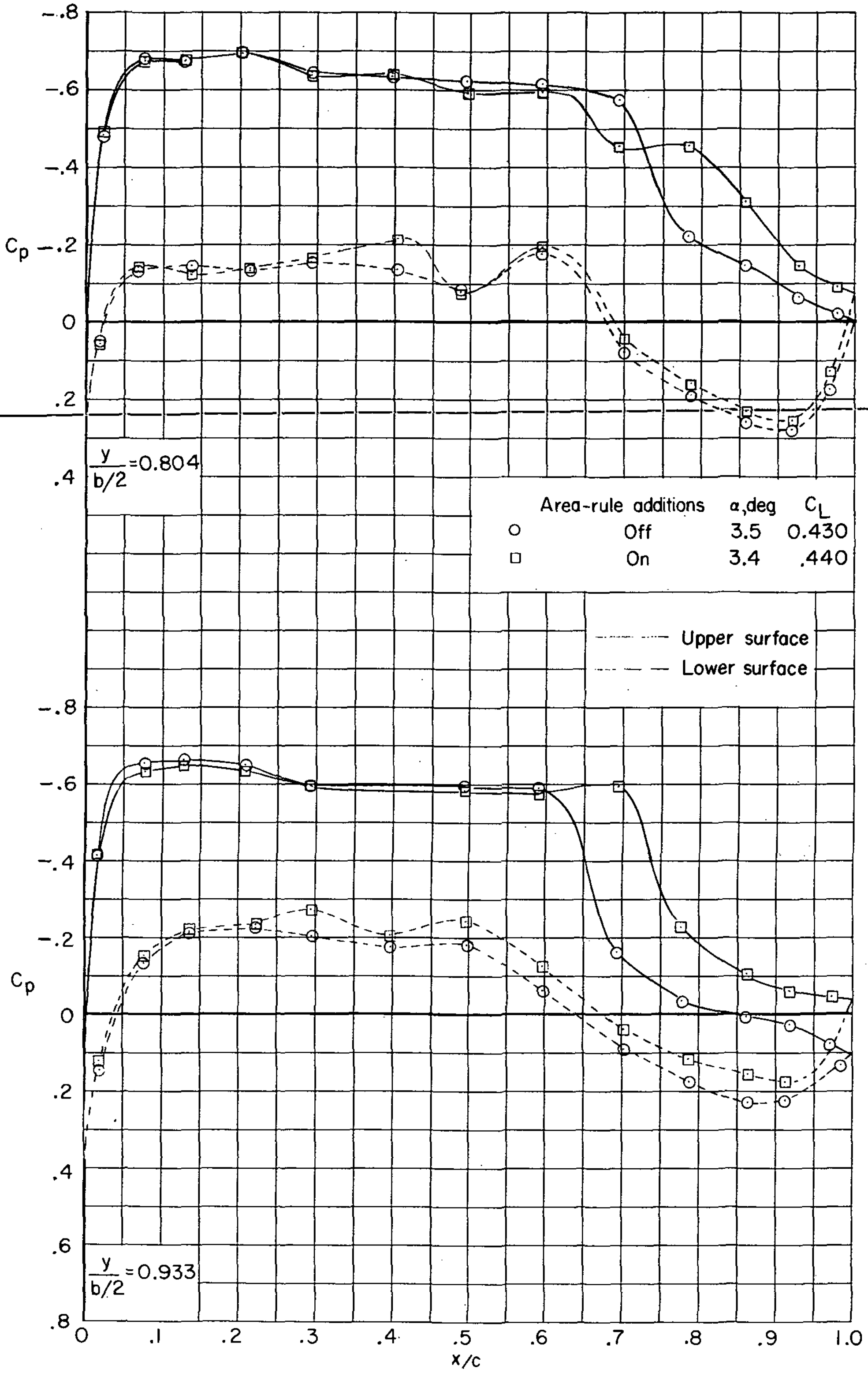
[redacted]



(c) M = 0.99. Continued.

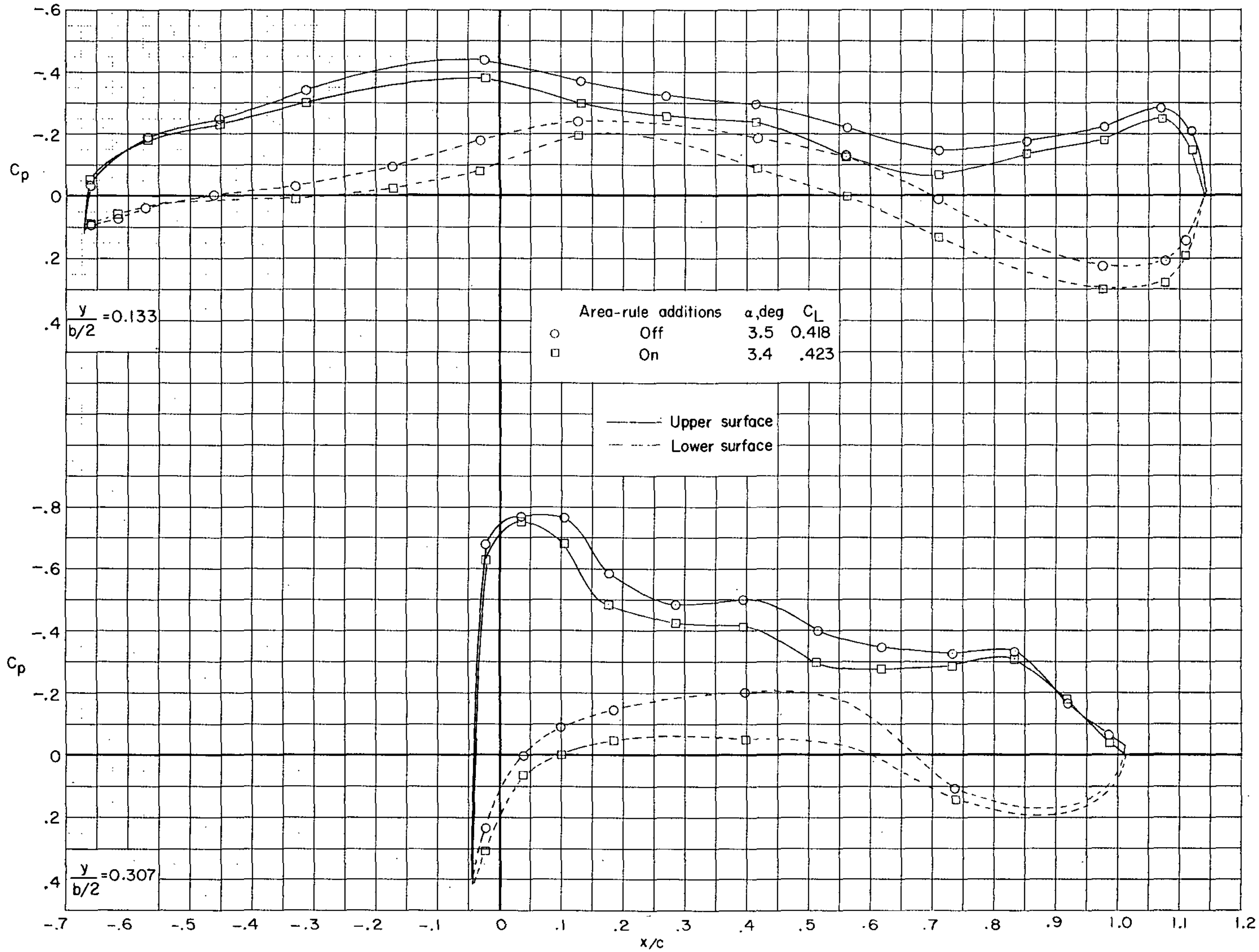
Figure 5.- Continued.

[redacted]



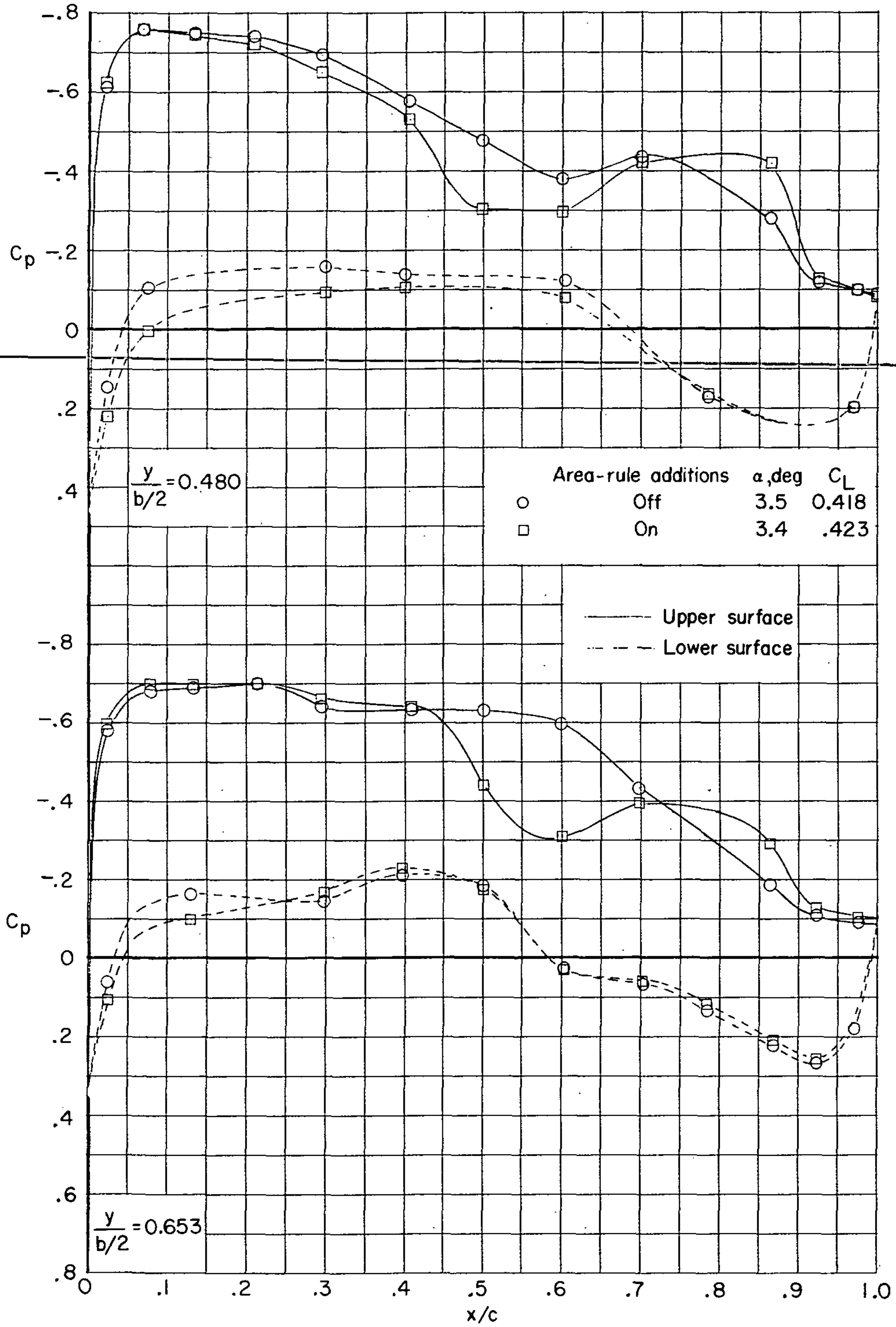
(c) M = 0.99. Concluded.

Figure 5.- Continued.



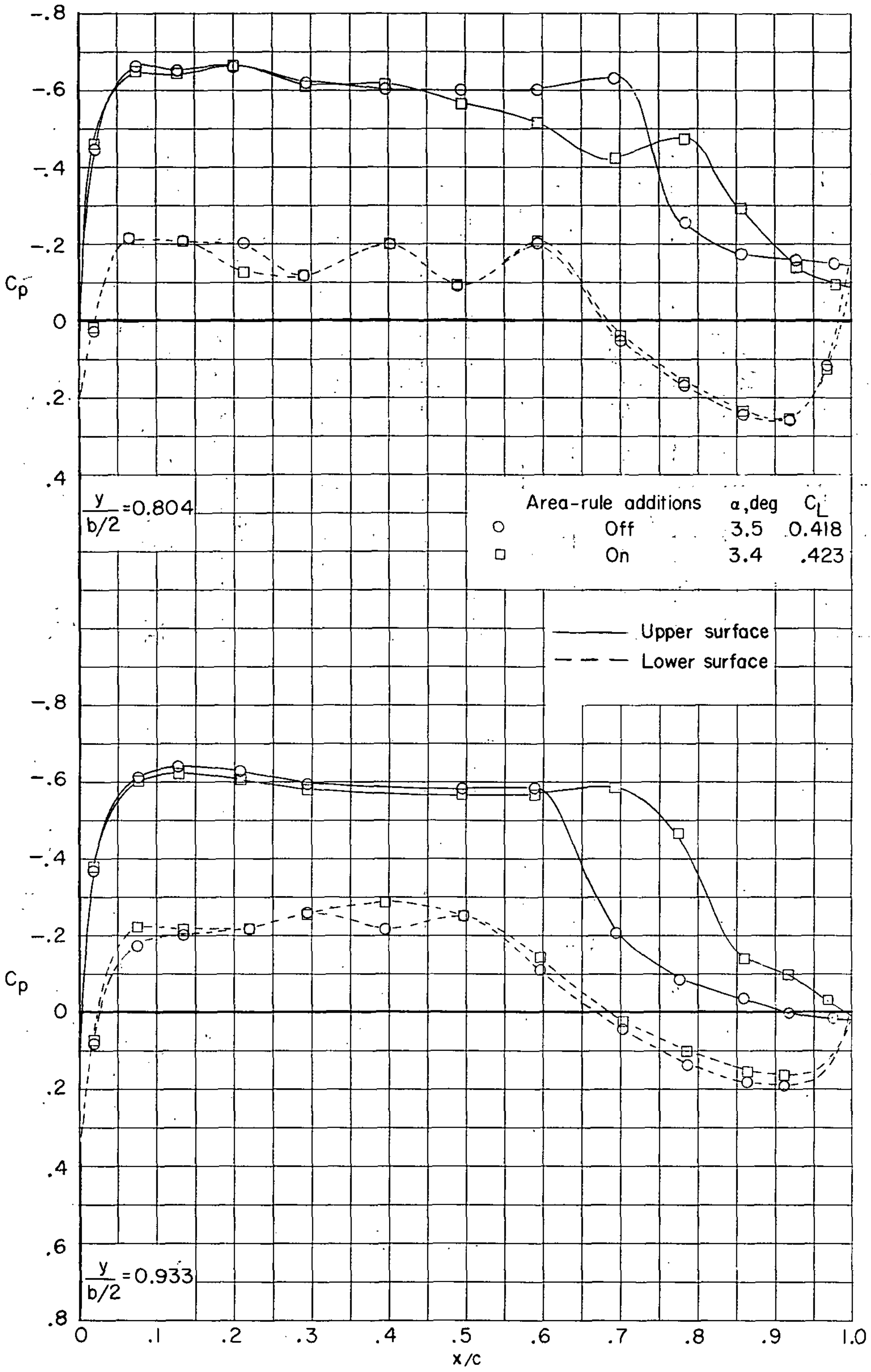
(d) M = 1.00.

Figure 5.- Continued.



(d) M = 1.00. Continued.

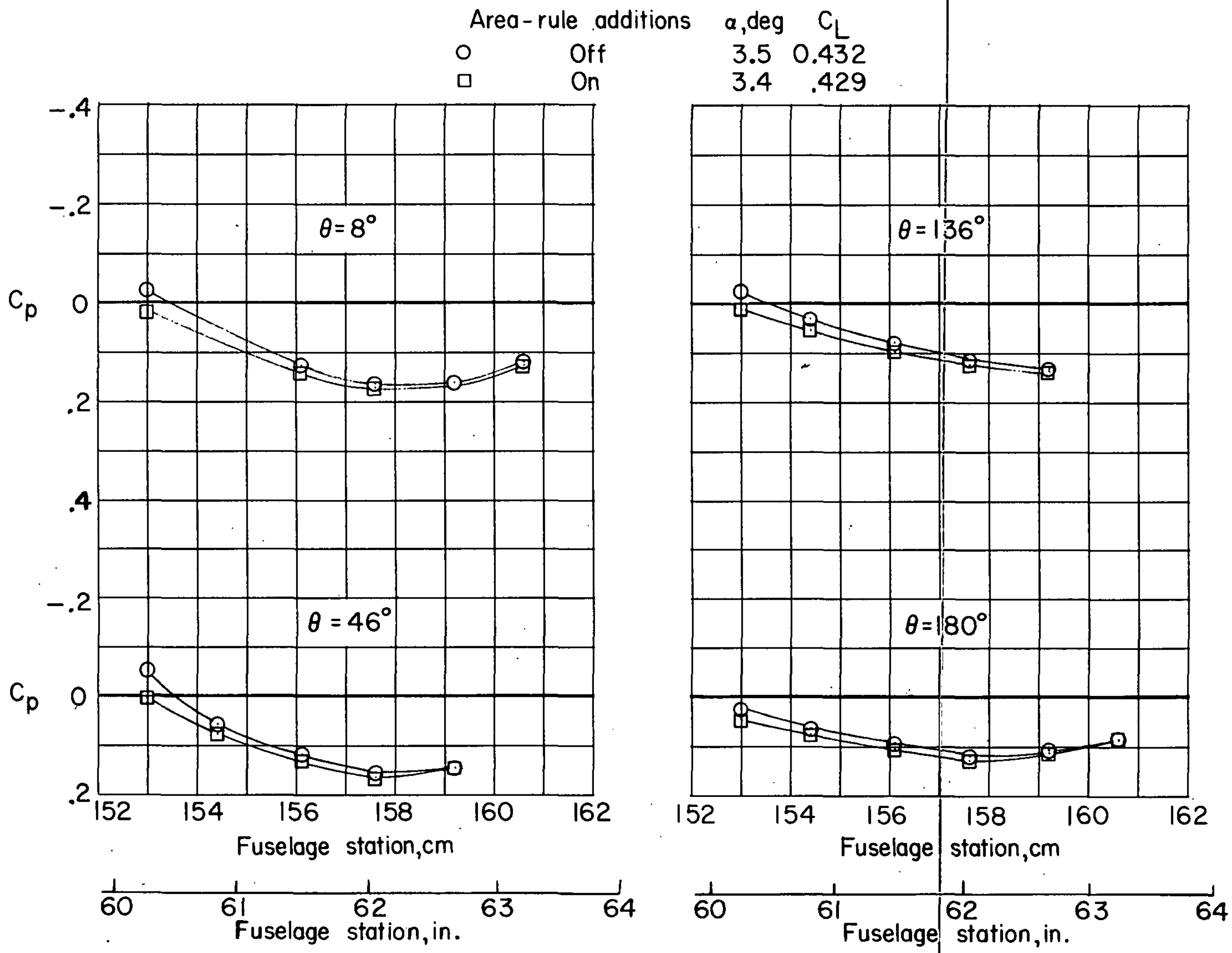
Figure 5.- Continued.



(d) M = 1.00. Concluded.

Figure 5.- Concluded.

[redacted]

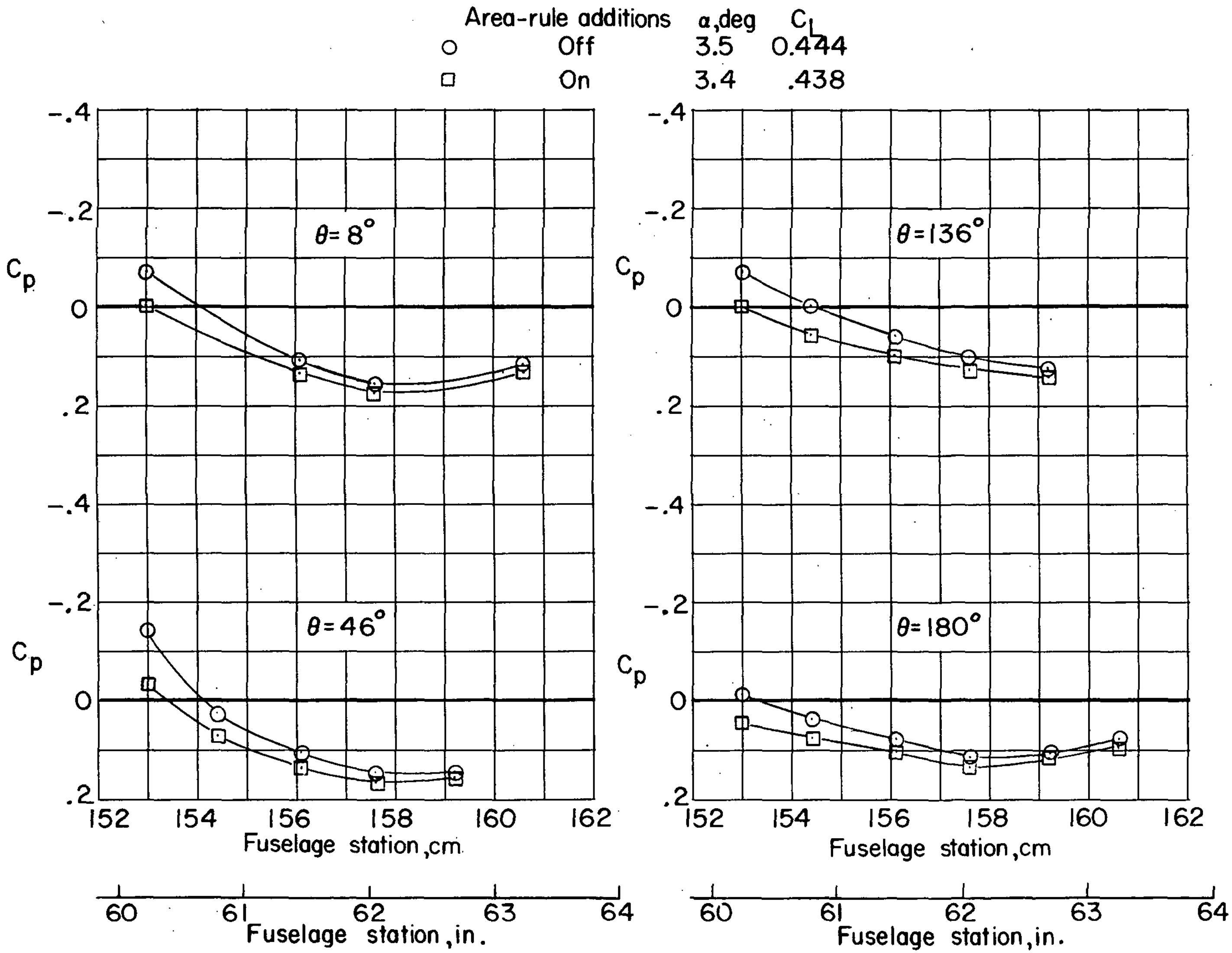


(a) M = 0.97.

Figure 6.- Effect of fuselage area-rule additions on pressure distributions over rear of fuselage near cruise lift coefficient. $\beta = 0^{\circ}$; $\delta_h = -2.5^{\circ}$.

[redacted]

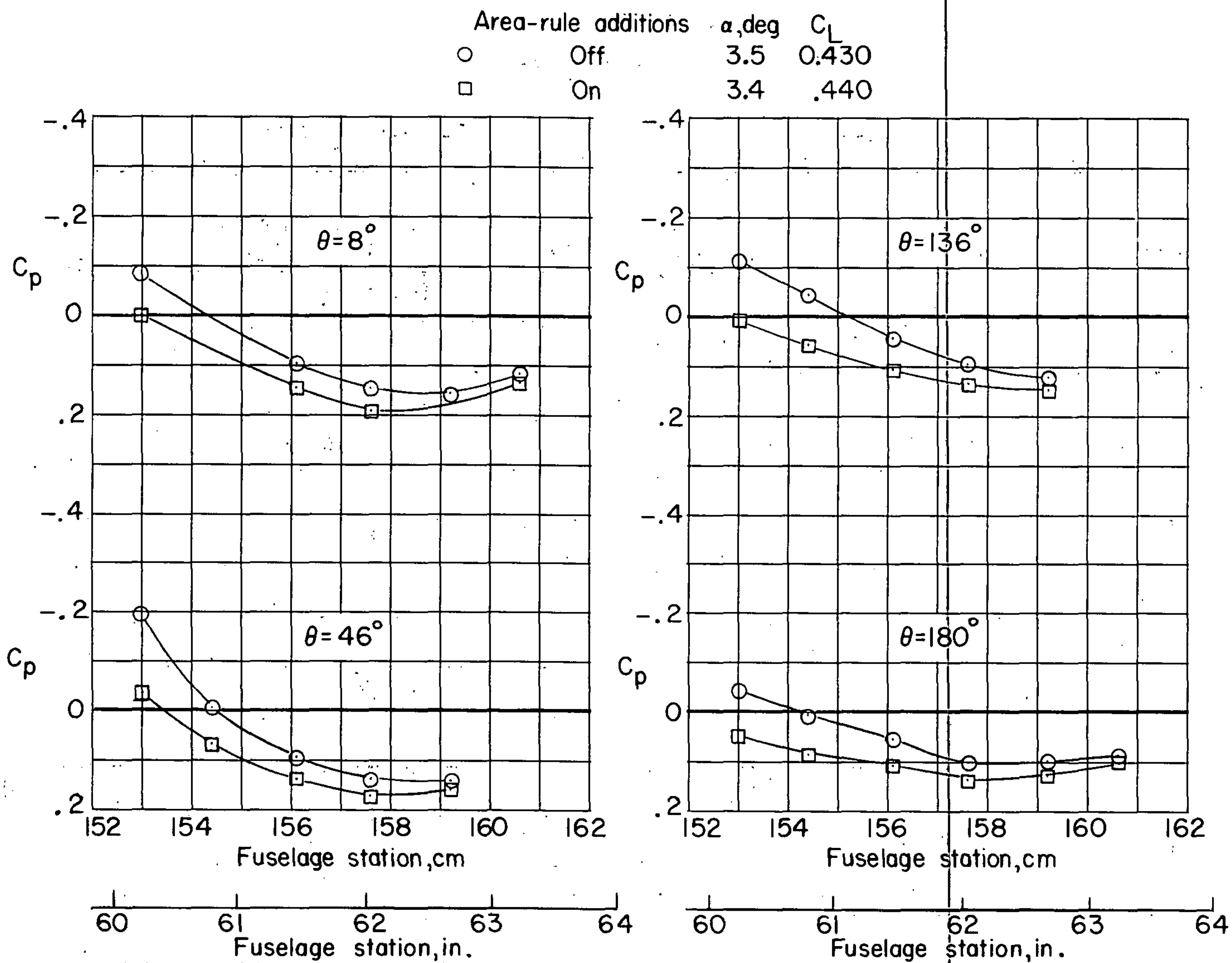
[redacted]



(b) M = 0.98.

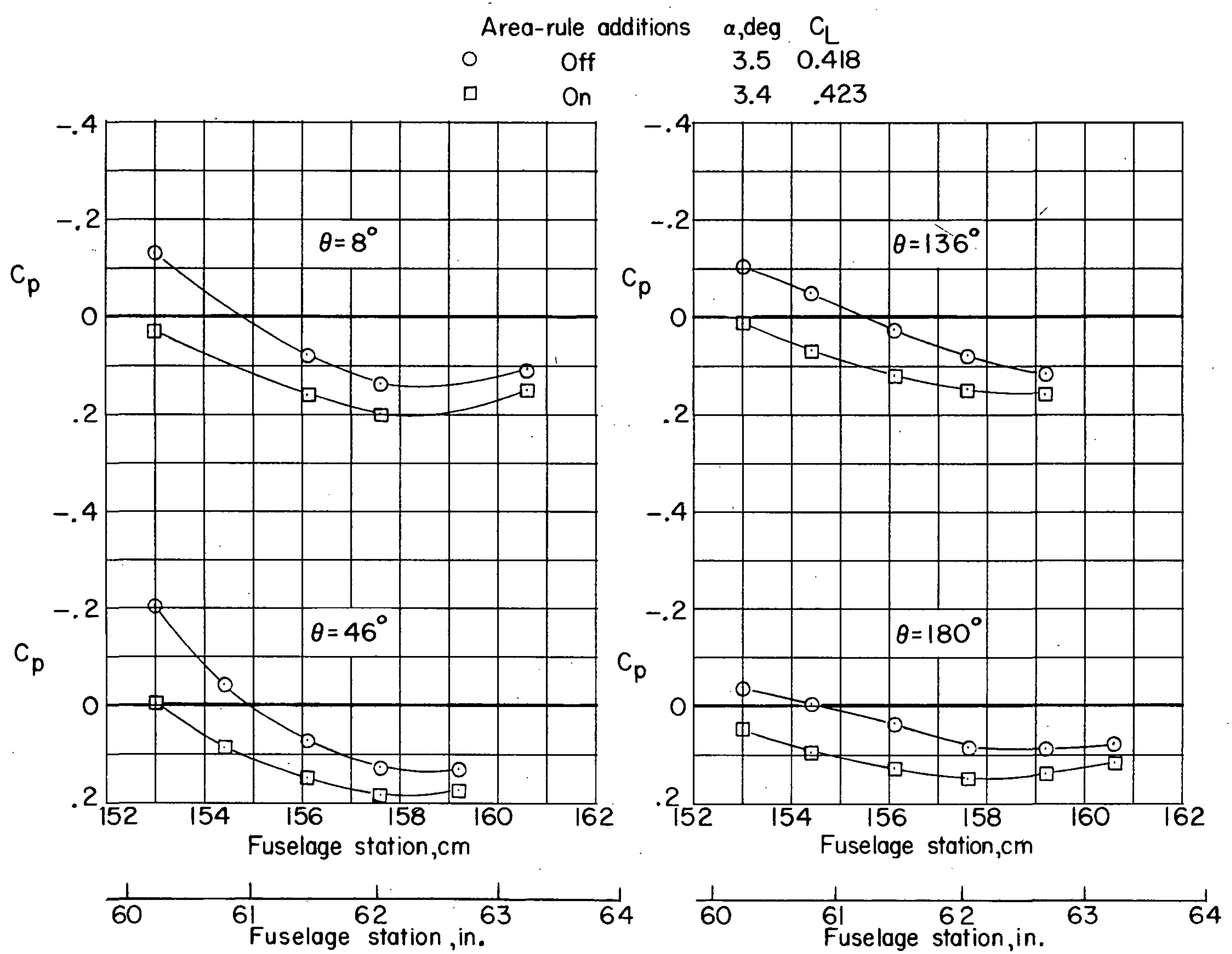
Figure 6.- Continued.

[redacted]



(c) M = 0.99.

Figure 6.- Continued.



(d) M = 1.00.

Figure 6.- Concluded.

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16. Abstract

Basic pressure measurements have been made on a 0.087-scale model of a supercritical-wing research airplane in the Langley 8-foot transonic pressure tunnel at Mach numbers from 0.25 to 1.00 to determine the effects on the local aerodynamic loads over the wing and rear fuselage of area-rule additions to the sides of the fuselage. In addition, pressure measurements over the surface of the area-rule additions themselves were obtained at angles of sideslip of approximately -5°, 0°, and 5° to aid in the structural design of the additions. Except for representative figures, results are presented in tabular form without analysis.

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To **UNCLASSIFIED**

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